

environmental, and energy aspects of the rule that might suggest a need to modify the rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report that summarizes each FAA-public contact concerned with the substance of this AD will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this action must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 2002-NE-45-AD." The postcard will be date stamped and returned to the commenter.

Regulatory Analysis

This final rule does not have federalism implications, as defined in Executive Order 13132, because it would not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Accordingly, the FAA has not consulted with state authorities prior to publication of this final rule.

The FAA has determined that this regulation is an emergency regulation that must be issued immediately to correct an unsafe condition in aircraft, and is not a "significant regulatory action" under Executive Order 12866. It has been determined further that this action involves an emergency regulation under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979). If it is determined that this emergency regulation otherwise would be significant under DOT Regulatory Policies and Procedures, a final regulatory evaluation will be prepared and placed in the Rules Docket. A copy of it, if filed, may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

2003-03-21 Pratt & Whitney Canada:

Amendment 39-13046. Docket No. 2002-NE-45-AD.

Applicability: This airworthiness directive (AD) is applicable to Pratt & Whitney Canada (PWC) PW530A, PW535A, and PW545A turbofan engines with flexible fuel tubes part numbers (P/N's) 30J2285-01, 3054416-01, and 30J2323-01 installed. These engines are installed on, but not limited to Cessna Citation model 550 "Bravo", model 560XL "Excel", and model 560 "Encore" airplanes.

Note 1: This AD applies to each engine identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For engines that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Compliance with this AD is required within 50 flight hours, but no later than 60 days after the effective date of this AD, whichever occurs first, unless already done.

To prevent a fire in the engine nacelle, do the following:

(a) Replace flexible fuel tubes P/N's 30J2285-01 (PW530A engines), 3054416-01 (PW535A engines), and 30J2323-01 (PW545A engines) with flexible fuel tubes P/N's 30J2578-01, 3058704-01, and 30J2579-01 respectively; or

(b) Perform a one-time visual external and internal inspection of flexible fuel tubes P/N's 30J2285-01 (PW530A engines), 3054416-01 (PW535A engines), and 30J2323-01 (PW545A engines), and fuel tube part number re-marking, in accordance with paragraphs 3.A.(1) through 3.A.(8) of the Accomplishment Instructions of PWC SB PW500-72-30217, Revision 1, dated July 29, 2002.

Alternative Methods of Compliance

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Engine Certification Office. Operators must submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Engine Certification Office.

Note 2: Information concerning the existence of approved alternative methods of compliance with this airworthiness directive, if any, may be obtained from the Engine Certification Office.

Special Flight Permits

(d) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be done.

Documents That Have Been Incorporated by Reference

(e) The tube inspection or replacement must be done in accordance with Pratt & Whitney Canada Service Bulletin PW500-72-30217, Revision 1, dated July 29, 2002. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Pratt & Whitney Canada, 1000 Marie-Victorin, Longueuil, Quebec, Canada J4G1A1. Copies may be inspected at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 3: The subject of this AD is addressed in Transport Canada airworthiness directive CF-2002-42, dated September 30, 2002.

Effective Date

(f) This amendment becomes effective on February 20, 2003.

Issued in Burlington, Massachusetts, on January 29, 2003.

Jay J. Pardee,

Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. 03-2632 Filed 2-4-03; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002-NM-240-AD; Amendment 39-13047; AD 2003-03-22]

RIN 2120-AA64

Airworthiness Directives; Boeing Model 737-600, -700, -700C, -800, and -900 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Boeing Model 737-600, -700, -700C, -800, and -900 series airplanes, that requires installing speedbrake limitation placards in the

flight compartment, and revising the Limitations Section of the Airplane Flight Manual to ensure the flightcrew is advised not to extend the speedbrake lever beyond the flight detent. For certain airplanes, this AD requires modifying the elevator and elevator tab assembly. This action is necessary to prevent severe vibration of the elevator and elevator tab assembly, which could result in severe damage to the horizontal stabilizer followed by possible loss of the elevator tab and consequent loss of controllability of the airplane. This action is intended to address the identified unsafe condition.

DATES: Effective March 12, 2003.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of March 12, 2003.

ADDRESSES: The service information referenced in this AD may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124-2207. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Nancy H. Marsh, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 917-6440; fax (425) 917-6590.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Boeing Model 737-600, -700, -700C, -800, and -900 series airplanes was published in the **Federal Register** on November 15, 2002 (67 FR 69157). That action proposed to require installing speedbrake limitation placards in the flight compartment, and revising the Limitations Section of the Airplane Flight Manual (AFM) to ensure the flightcrew is advised not to extend the speedbrake lever beyond the flight detent. For certain airplanes, that action also proposed to require modifying the elevator and elevator tab assembly.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comments received. One commenter concurs with the proposed AD. Two

commenters state that they have no comments on the proposed AD.

Request To Clarify Certain Language in Paragraph (a)(2)

One commenter asks that certain language, as specified in paragraph (a)(2) of the proposed AD, be clarified. The commenter notes that the current language, which is to be included in the AFM, states the following: "Do not extend the speedbrake lever beyond the flight detent in flight." That statement, as written, does not match the language specified in the existing AFM. The language should be changed, for clarification, to match the AFM language and should state: "In flight, do not extend the speedbrake lever beyond the FLIGHT detent."

The FAA agrees with the commenter in that the language used in paragraph (a)(2) of the AD should be clarified to match the AFM language. We have changed paragraph (a)(2) of this final rule accordingly.

Request To Change Discussion Section

The same commenter asks that the Discussion section of the proposed AD be changed to remove the reference to Boeing Model 737-900 series airplanes from the first sentence. That sentence states, "The FAA has received several reports of excessive in-flight vibrations of the elevator and elevator tab on certain Boeing Model 737-600, -700, -700C, -800, and -900 series airplanes." The commenter notes that no excessive in-flight vibrations of the elevator and elevator tab have occurred on Model 737-900 series airplanes.

The same commenter asks that certain terminology in the Discussion section of the proposed AD be changed. That section reads, in part, "[t]he elevator and elevator tab are susceptible to excessive vibration and, under certain conditions, limit-cycle flutter. These vibration events have been attributed to loose or missing components, excessive wear, or excessive freeplay of the tab." The commenter requests that it be changed to, "[t]he elevator and elevator tab are susceptible to excessive vibration and, under certain conditions, limit-cycle oscillation (LCO). These vibration events have been attributed to lack of torsional rigidity (in the case of LCO); or missing components, excessive wear, or excessive freeplay of the tab." The commenter states that LCO is the accepted and proper term to use when referring to the severe vibrations associated with lack of torsional stiffness.

We acknowledge that no excessive in-flight vibrations of the elevator and elevator tab have been reported on

Model 737-900 series airplanes in-service. The intent of the Discussion section is to provide the background and events that prompted the proposed AD, and to specify that vibrations did occur on Model 737-600, -700, -700C, and -800 series airplanes in-service.

We acknowledge that the term "lack of torsional rigidity" is a valid term and could be used to describe a design deficiency that also contributes to excessive in-flight vibration. However, the terms "LCO" and "LCF" are not commonly used terms in the airline industry; these terms are used primarily by airplane manufacturers. We have concluded that the term "high amplitude oscillations of the elevator tab" best describes the condition in a manner understood by the airline industry.

Since the Discussion section of a proposed AD is not restated in a final rule, no change to this final rule is necessary to address the issues raised by the commenters.

Conclusion

After careful review of the available data, including the comments noted above, the FAA has determined that air safety and the public interest require the adoption of the rule with the change previously described. The FAA has determined that this change will neither increase the economic burden on any operator nor increase the scope of the AD.

Cost Impact

There are approximately 1,174 airplanes of the affected design in the worldwide fleet. We estimate that 550 airplanes of U.S. registry will be affected by this AD.

It will take approximately 1 work hour per airplane to accomplish the required placard installation, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the required installation on U.S. operators is estimated to be \$33,000, or \$60 per airplane.

It will take approximately 1 work hour per airplane to accomplish the required AFM revision, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the required revision on U.S. operators is estimated to be \$33,000, or \$60 per airplane.

It will take approximately 88 work hours per airplane to accomplish the required modification of the elevator and elevator tab assembly, at an average labor rate of \$60 per work hour. The FAA has been advised by Boeing that the manufacturer will provide parts for the elevator/tab retrofit, including

shipping, at no cost to operators. The manufacturer will have operators "exchange" their existing parts for new parts to support the retrofit program. Based on this information, the cost impact of the required modification on U.S. operators is estimated to be \$2,904,000, or \$5,280 per airplane.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

2003-03-22 Boeing: Amendment 39-13047. Docket 2002-NM-240-AD.

Applicability: Model 737-600, -700, -700C, -800, and -900 series airplanes; line numbers 1 through 1174 inclusive; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent severe vibration of the elevator and elevator tab assembly, which could result in severe damage to the horizontal stabilizer followed by possible loss of the elevator tab and consequent loss of controllability of the airplane, accomplish the following:

Airplane Flight Manual (AFM) Revision/Placard Installation

(a) For Model 737-600, -700, -700C, -800, and -900 series airplanes having line numbers 1 through 1043 inclusive: Within 90 days after the effective date of this AD, do the actions specified in paragraphs (a)(1) and (a)(2) of this AD.

(1) Install a speedbrake limitation placard on the P1-1 and P3-3 panel assemblies per Figure 1 or Figure 2, as applicable, of paragraph 3.B., "Work Instructions," of the Accomplishment Instructions of Boeing Alert Service Bulletin 737-11A1109, dated March 28, 2002.

(2) Revise the Limitations Section of the FAA-approved AFM to include the following statement (this may be accomplished by inserting a copy of this AD in the AFM): "In flight, do not extend the speedbrake lever beyond the FLIGHT detent."

Modification

(b) For Model 737-600, -700, -700C, and -800 series airplanes having line numbers 1 through 1174 inclusive: Before the accumulation of 18,000 total flight cycles, or within 2 years after the effective date of this AD, whichever occurs first, modify the elevator and elevator tab assemblies

(including installation of a new clevis fitting and a new tab mechanism on the horizontal stabilizer and, for certain airplanes, examination of the hinge plates on the stabilizer trailing edge to make sure the specified hinges are installed; changes to the seals in the balance bays; and installation of new elevators and tab assemblies, followed by adjustments and tests of the new installation), per the Accomplishment Instructions of Boeing Alert Service Bulletin 737-55A1080, dated September 19, 2002.

(c) Accomplishment of the modification required by paragraph (b) of this AD terminates the actions required by the ADs specified in the following table:

AD Number	Amendment Number
AD 99-15-09	39-11229
AD 99-18-01	39-11267
AD 2001-08-09	39-12186
AD 2001-09-51	39-12251
AD 2001-12-51	39-12294
AD 2001-14-05	39-12315
AD 2002-08-20	39-12732
AD 2002-08-52	39-12727

Operator's Equivalent Procedure

(d) If the Accomplishment Instructions of Boeing Alert Service Bulletin 737-55A1080, dated September 19, 2002, specify that the actions may be accomplished in accordance with an operator's "equivalent procedure:" The actions must be accomplished per the applicable chapter of the Boeing 737 Airplane Maintenance Manual specified in the alert service bulletin.

Alternative Methods of Compliance

(e) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

Special Flight Permit

(f) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(g) Unless otherwise specified in this AD, the actions shall be done in accordance with Boeing Alert Service Bulletin 737-11A1109, dated March 28, 2002; and Boeing Alert Service Bulletin 737-55A1080, dated September 19, 2002; as applicable. This incorporation by reference was approved by the Director of the **FEDERAL REGISTER** in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124-2207. Copies may be inspected at the FAA, Transport Airplane

Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Effective Date

(h) This amendment becomes effective on March 12, 2003.

Issued in Renton, Washington, on January 29, 2003.

Ali Bahrami,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-2496 Filed 2-4-03; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2003-CE-07-AD; Amendment 39-13043; AD 2003-03-18]

RIN 2120-AA64

Airworthiness Directives; Raytheon Aircraft Company Beech Models 1900, 1900C, and 1900D Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment adopts a new airworthiness directive (AD) that applies to all Raytheon Aircraft Company (Raytheon) Beech Models 1900, 1900C, and 1900D airplanes. This AD requires you to perform control column sweep and stop bolt inspections to verify full elevator travel to the primary up and down stops and that the stop bolt length is not excessive, re-rig the elevator control system if the airplane does not pass the control column sweep and stop inspections, and do a more detailed inspection at a later time if the airplane does pass the inspection. This AD also requires you to report the results of certain inspections. This AD is the result of recent ground testing and a review of the rigging procedures of a Raytheon Beech Model 1900D airplane, which reveals that the elevator control system could be mis-rigged to restrict elevator travel if current maintenance procedures are not properly followed. In these instances, it may appear to the crew that they have full elevator control column movement. However, the elevator may not have full travel. Such restricted travel may remain undetected until the airplane is operated in a loading condition that requires full elevator authority to control the pitch. The actions specified by this AD are intended to detect and

correct any mis-rigged elevator control system, which could lead to insufficient elevator control authority and loss of control of the airplane.

DATES: The AD becomes effective February 5, 2003, to all affected persons who did not receive emergency AD 2003-03-18, issued January 27, 2003. Emergency AD 2003-03-18 contained the requirements of this amendment and became effective immediately upon receipt and required the actions 4 days after issuance (January 31, 2003).

The Federal Aviation Administration (FAA) must receive any comments on this rule on or before March 7, 2003.

ADDRESSES: Submit comments to FAA, Central Region, Office of the Regional Counsel, Attention: Rules Docket No. 2003-CE-07-AD, 901 Locust, Room 506, Kansas City, Missouri 64106. You may view any comments at this location between 8 a.m. and 4 p.m., Monday through Friday, except Federal holidays. You may also send comments electronically to the following address: 9-ACE-7-Docket@faa.gov. Comments sent electronically must contain "Docket No. 2003-CE-07-AD" in the subject line. If you send comments electronically as attached electronic files, the files must be formatted in Microsoft Word 97 for Windows or ASCII text.

You may view information related to this AD at FAA, Central Region, Office of the Regional Counsel, Attention: Rules Docket No. 2003-CE-07-AD, 901 Locust, Room 506, Kansas City, Missouri 64106; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Paul DeVore, Aerospace Engineer, FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Mid-Continent Airport, Wichita, Kansas 67209; telephone: (316) 946-4142; facsimile: (316) 946-4407.

SUPPLEMENTARY INFORMATION:

Discussion

What Has Happened so Far?

Recent ground testing and a review of the rigging procedures of a Raytheon Beech Model 1900D airplane reveals that the elevator control system could be mis-rigged to restrict elevator travel if current maintenance procedures are not properly followed. In these instances, it may appear to the crew that they have full elevator control column movement. However, the elevator may not have full travel. Such restricted travel may remain undetected until the airplane is operated in a loading condition that requires full elevator authority to control the pitch.

The Raytheon Beech Models 1900 and 1900C airplanes incorporate the same elevator control system design and are affected by this condition.

In certain loading conditions, a mis-rigged elevator control system, if not detected and corrected, could lead to insufficient elevator control authority and loss of control of the airplane.

Raytheon has not issued service information regarding this subject. Rigging procedures are included in the applicable Raytheon 1900/1900C or 1900D maintenance manual.

On January 27, 2003, FAA issued emergency AD 2003-03-18 to require you to:

- Perform control column sweep and stop bolt inspections to verify full elevator travel to the primary up and down stops and to verify that the stop bolt length is not excessive;

- If the airplane does not pass the initial control column sweep and stop bolt inspections, re-rig and/or do a more detailed inspection of the elevator control system;

- If the airplane does pass the initial control column sweep and stop bolt length inspections, do a more detailed inspection within 100 hours time-in-service (TIS); and

- Report the results of the initial inspection and the 100-hour TIS inspection (if applicable).

Why Is it Important to Publish This AD?

The FAA found that immediate corrective action was required, that notice and opportunity for prior public comment were impracticable and contrary to the public interest, and that good cause existed to make the AD effective immediately by individual letters issued on January 27, 2003, to all known U.S. operators of Raytheon Beech Models 1900, 1900C, and 1900D airplanes. These conditions still exist, and the AD is published in the **Federal Register** as an amendment to section 39.13 of the Federal Aviation Regulations (14 CFR 39.13) to make it effective to all persons.

Comments Invited

How Do I Comment on This AD?

Although this action is in the form of a final rule and was not preceded by notice and opportunity for public comment, FAA invites your comments on the rule. You may submit whatever written data, views, or arguments you choose. You need to include the rule's docket number and submit your comments to the address specified under the caption **ADDRESSES**. We will consider all comments received on or before the closing date specified above.