NASA Technical Memorandum 86683

Pressure-Distribution Measurements on a Transonic Low-Aspect Ratio Wing

Earl R. Keener

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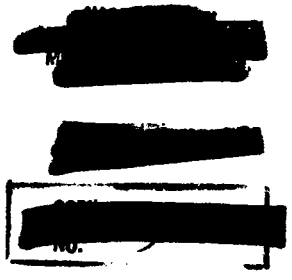

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Earl R. Keener, Ames Research Center, Moffett Field, California

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NOMENCLATURE

Symbol	Computer Symbol	Definition
a	ALPHA	angle of attack, deg
b/2	B/2	wing semispan
c		local wing chord
c_{av}		average wing chord, S/b
$\bar{c}$	MAC	wing mean aerodynamic chord, $(2/S)\int_0^1 c^2\, d(2y/b)$
	CONF	configuration identification number
c_r	CR	root chord
c_t		tip chord
C_p	CP	pressure coefficient, $(p - p_\infty)/q$
M	MACH	free-stream Mach number
n	n	nondimensional spanwise distance from wing root, 2y/b
p	P	free-stream static pressure, psf
p_t	PT	free-stream total pressure, psf
q	Q	free-strem dynamic pressure, psf
$Re/\bar{c}$	RN/L	free-stream unit Reynolds number, M per ft
Re	RN	Reynolds number based on $\bar{c}$, M
S/2		area of semispan wing model
t_t	TTR	free-stream total temperature, °R
x	X	chordwise distance rearward of leading edge

x'		chordwise distance from $0.25\bar{c}$ (wing pitching-moment axis) to $0.25c$ (section pitching-moment axis), $0.9559(c - \bar{c})$
y	Y	spanwise distance outboard of wing root

Wing Section Aerodynamic Characteristics

c_n	CNS	wing-section normal-force coefficient
	CNC/	section normal-load parameter, $c_n(c/c_{av})$
c_m	CMS	wing-section pitching-moment coefficient about $c/4$
c'_m		wing-section pitching-moment coefficient about the wing pitching-moment axis passing through $0.25\bar{c}$, $c_m + (x'/\bar{c})c_n$, used in C_M
	CMC/	section pitching-moment parameter, $c'_m(c/c_{av})^2$
x_{cp}	XCPS	wing-section chordwise center of pressure, % chord

Wing Aerodynamics Characteristics

C_B	CB	wing bending-moment coefficient; moment axis is wing root chord
C_N	CN	wing normal force coefficient
C_M	CM	wing pitching-moment coefficient; moment axis passes through $0.25\bar{c}$, see c'_m
X_{CP}	XCP	wing chordwise center of pressure, % mean-aerodynamic chord
Y_{CP}	YCP	wing spanwise center of pressure, % semispan

Subscripts

L	( )L	lower surface
U	( )U	upper surface

SUMMARY

Surface-pressure distributions and oil-flow photographs are presented from wind-tunnel tests of a large-scale (0.90 m) semispan model of NASA/Lockheed Wing C, a generic transonic, supercritical, low-aspect-ratio, highly three-dimensional (3-D) configuration, designed to conduct 3-D boundary-layer tests. The wing was designed using a 3-D, transonic, full-potential-flow wing code (FLO22) and an optimization routine. Tests were conducted at the design angle of attack of 5° over a Mach number range from 0.25 to 0.96, and a Reynolds number range of 3.4×10^6 to 10×10^6. Pressures were measured with the suction slots of the tunnel floor and ceiling open for most of the tests but taped closed for some tests to simulate solid walls. This paper presents the surface-pressure measurements and the oil-flow patterns, obtained to determine the extent of 3-D surface flow in preparation for the boundary-layer measurements. A comparison is made with pressures from a small-scale model tested at the same Reynolds number in a high Reynolds number facility by Lockheed-Georgia Company and with predicted pressures using two 3-D, full-potential-flow, transonic wing codes: design code FLO22 (nonconservative) and TWING code (conservative).

At the design Mach number and angle of attack of 0.85 and 5°, the most prominent features in the oil-flow patterns were the unexpected local-flow separation that occurred in the outer 30% of the semispan and the lack of 3-D boundary-layer flow over the rest of the wing. The local separation was caused by a strong, local, shock-wave/boundary-layer interaction that was not a tip vortex effect. The flow separation increased as the Mach number was increased to 0.95, but disappeared when the Mach number was reduced to 0.82 where the surface oil-flow angles were less than 10° over most of the wing. The main wing shock wave was unsteady at a low, irregular frequency of ~3 Hz, inducing unsteady pressures to the trailing edge.

Comparing large-scale and small-scale data from two wind tunnels with each other and with predictions can be difficult due to wall interference and model boundary-layer effects. The comparisons herein show that the method of matching leading-edge pressures appears to be one satisfactory way of selecting the experimental angle of attack to correlate the experimental and predicted pressure distributions. Using this method, predictions by FLO22 and TWING codes agree rather well with each other and with the experiments, except for small variations in shock position and aft loading. It is shown that the flow separation at the design conditions might have been avoided by further iteration in the design.

Wall-interference effect was effectively demonstrated when the floor and ceiling suction slots were taped closed to simulate solid walls, and the normal-force coefficient increased tremendously from 0.52 to 0.65.

The lack of 3D boundary-layer flow on Wing C raised the question: under what design conditions are wing boundary layers significantly 3-D for unseparated flow? Evidence presented from this study and from other cited wing studies indicate that wings that are optimized for mild shock waves and mild pressure-recover gradients generally have small 3-D boundary layer flow (flow angles less than 10°) at design conditions for unseparated flow. Additional evidence from another cited wing study indicates that in some wing designs the optimization is relaxed to allow the boundary layer to approach separation at the design conditions, which induces significant 3-D boundary-layer flows near the trailing edge.

INTRODUCTION

In recent years, significant advancements have been made in computational methods for the design and analysis of transonic flow about wings. There is a continuing requirement, however, to assess the accuracy and efficiency of current computational methods by systematic comparisons with reliable experimental data. To contribute to the current efforts to validate existing inviscid and viscous numerical codes, the Aeronautics Research Branch of Ames Research Center (ARC) engaged in two cooperative studies of several wing models to obtain 3-dimensional (3-D) pressure distributions and boundary-layer data. This paper presents results from the ARC contribution to the first study and contrasts these results with those of the second study.

In the first study with the Lockheed-Georgia Company, a cooperative computational-experimental investigation was conducted to obtain pressure-distributions and 3-D boundary-layer data on a generic model of a modern, highly three-dimensional, advanced-technology wing configuration. The wing was designed using a 3-D, nonconservative, full-potential-flow, transonic wing code (FLO22) and an optimization routine. A highly swept, low-aspect ratio wing was selected that had supercritical airfoils with relatively thick sections, moderate aft loading, mild shock waves, and a mild pressure recovery. The result was a highly optimized wing (designated Wing C), designed for unseparated flow at a design Mach number of 0.85 and a design lift coefficient of about 0.5 at an angle of attack of about 5°. A small-scale semispan model of Wing C was tested by Lockheed-Georgia in their high Reynolds number facility (the Compressible Flow Wing Tunnel (CFWT)) at a Reynolds number of 10 million, based on the mean aerodynamic chord. In addition, two other small-scale models were designed and tested: a transport-type wing and a fighter-type wing (designated Wings A and B). Surface pressures were measured on the wing and on the tunnel walls for comparison with calculations of wall effects on the boundary conditions from Computational Fluid Dynamics (CFD) codes. Hinson and Burdges published both the small-scale data in reference 1 and a comparison of the small-scale measurements with several 3-D transonic inviscid codes in references 2 and 3. Lemmerman and Atta published predictions of the boundary-layer thickness and skin friction, made with several 3-D transonic boundary-layer codes in reference 4.

This paper presents the results from the ARC contribution to the Lockheed Georgia cooperative program: tests of a large-scale (0.90 m) semispan model of low-aspect-ratio Wing C, built to obtain thick boundary layers for ease of measurement in a large wind tunnel (the Ames 6- by 6-ft Transonic/Supersonic Wind Tunnel). Surface-pressure measurements, oil-flow studies, and boundary-layer surveys were obtained at several wing stations at the design angle of attack of 5° over a Mach number range of 0.25 to 0.96 and a Reynolds number range of 3.4×10^6 to 10×10^6. Wing pressures were measured with the tunnel floor and ceiling suction slots open for most of the tests but then taped closed for some tests to simulate solid walls for comparison with predictions of tunnel-wall effect. The measured pressures are compared with the small-scale wing pressures and with the predictions from two 3-D, full-potential-flow, transonic wing codes: design code FL022 (nonconservative) and TWING code (conservative). Selected measurements and computations of surface-pressure distributions and photographs of oil-flow tests were published in reference 5.

Although a number of computational-experimental comparisons have been made in recent years, the present test results are enhanced by results from two different models in two different wind tunnels. The major objectives of the discussion are to consider the extent of 3-D boundary-layer flow at the design condition (as indicated by the oil-flow tests), in preparation for the boundary-layer tests; the cause of the unexpected occurrence of local-flow separation at the design condition; the effects of tunnel-wall interference on the effective lift and Mach number of the two models tested in the two tunnels; and the general success of the predictions of the pressure distributions.

In the second afore-mentioned cooperative study, Spaid of McDonnell Douglas Research Laboratory, thoroughly investigated the boundary-layer characteristics of a semispan wing-transport model (ref. 6). The resulting combined research effort of the cooperative studies (refs. 3 and 6 and the present results) constitutes a substantial contribution to the data base and the analysis of computational fluid dynamics: i.e., data from three widely different small-scale models, data from two different-size models tested in two different wind tunnels, data for both small and large 3-D boundary-layer flow, and data for both unseparated flow and for shock-wave/boundary-layer separated flow.

The author wishes to acknowledge the substantial efforts of B. L. Hinson, K. P. Burdges, and L. A. Lemmerman of Lockheed-Georgia Comapny in the design of the wing and the contribution to the cooperative test planning; to the NASA Model Development Branch, J. Peterson, Chief, for supervising the building of an outstanding model; to D. Penna, CALSPAN, Inc., for his extensive contribution as project engineer in charge of the wind-tunnel test; to M. Wright, CALSPAN, Inc., for an extensive computer program, to G. Reynolds, CALSPAN, Inc., for the excellent oil-flow photography; to Informatics, Inc., (Joan Thomson and others), for computations and plot designs; and to Raymond Hicks, Ames Research Scientist and Computational Fluid Dynmaic Specialist, for his important contributions to the design of the wing and to the analysis of the comparisons of experiment and predictions.

WING DESIGN

Figure 1 shows planform sketches of the three wings, A, B, and C, which were designed for the Ames/Lockheed-Georgia Company cooperative computational/ experimental investigation of transonic wing-design technology. Wing A was intended to represent a high-aspect-ratio transport wing, and Wing B a moderate-aspect-ratio fighter wing. Wing C is a generic design, not intended to represent any existing full-scale wing.

Wing C (fig. 2), the subject of the present study, was designed for the cooperative research program by R. Hicks and B. Hinson, aerodynamic computational specialists for Ames and Lockheed Georgia, respectively. (Refer to ref. 7 for a discussion of some examples of successes and failures of transonic potential-flow codes.) Wing C is a highly 3-D low-aspect-ratio configuration, selected to be consistent with the test requirements that the wing have a large leading-edge sweep angle (45°) and a large mean-chord length to develop a thick, more easily measured boundary layer. It was decided not to design specifically for a strong 3-D flow but to optimize the design for a moderate aft loading, mild shock waves, and a mild pressure recovery. However, it was felt that a strong 3-D boundary layer would result by selecting a highly swept, low-aspect-ratio wing. The design condition selected was a Mach number of 0.85, and a lift coefficient of about 0.5, occurring at an angle of attack of 5°.

Two existing computer codes were used: FLO22 (ref. 8), an aerodynamic analysis program based on a relaxation solution of the 3-D, full-potnetial-flow equation, and a numerical optimization program based on the method of feasible directions (ref. 9). The FLO22 code was developed for analyzing inviscid, isentropic, transonic flow past 3-D swept wings. Weak shock waves are automatically located whenever they occur.

The design of the wing sections began by specifying the desired pressure distributions. For simplicity of construction, only two design control stations were selected ($n = 2y/b = 0.065$ and 0.91) so that linear lofting could be used between the root and tip stations. The specified design pressure distributions (shown for Wing C in fig. 3) were chosen to produce two objectives: a mild shock-wave pressure recovery from leading-edge suction pressures (accomplished by limiting leading-edge local Mach numbers to the commonly accepted maximum value of 1.2 normal to the leading edge for no flow separation), and a mild pressure recovery behind the shock wave to the trailing edge (which gave a moderate aft loading). The airfoil shapes were allowed to vary at the two control stations by minimizing the RMS deviation between the computed and the pre-selected design pressure distributions by appropriate modifications to the wing geometry, including chordwise camber and spanwise twist, using the FLO22 transonic solutions. Estimates of boundary-layer displacement effects, made by Lockheed/Georgia with an explicit-formulation 3-D code, indicated that the boundary layer did not significantly affect the design pressure distributions. The pressure distributions appeared to the designers to be well behaved. The larger percent thickness of the tip airfoil (10%), compared to the

root chord (6%), appeared to be acceptable and was retained (fig. 2). Final theoretical root and tip airfoil coordinates for Wing C are listed in table 1.

Typical calculated inviscid characteristics of the final design plotted using computer graphics programs developed at Ames for FL022 code are shown in figures 4 to 7. Figure 4 shows carpet plots of chordwise pressure distributions, and selected chordwise pressure distributions covering the range of $n = 0.73$ to 0.93. Figure 5 shows surface plots of velocity vectors, streamlines, isobars, and Mach-number contours. Figure 6 presents plots from a numerical chordwise cut of the flow field in a vertical plane at the midsemispan station showing flow field grid lines, pressure contours, Mach-number contours, and density contours. Computations at other span stations can be plotted. Figure 7 presents spanwise plots of load distribution and pitching-moment distribution.

TEST FACILITY

The Ames 6- by 6-Foot Transonic/Supersonic Wind Tunnel was chosen because the allowable model size and the tunnel operational characteristics were suitable for boundary-layer research. The tunnel is a variable pressure, continuous flow facility. The nozzle leading to the test section is of the asymmetric sliding-block type that permits a continuous variation of Mach number from 0.25 to 2.3. The test section has a slotted floor and ceiling with 6% porosity with provisions for boundary-layer removal. The turbulence-velocity level is high, measured to be about 1.5% of the free-stream velocity.

MODEL DESCRIPTION

A semispan (reflection plane) wing model was designed to be mounted on the tunnel wall because of its convenience and access to the instrumentation (fig. 8). Wing-root flow disturbances were not felt to be a problem because the flow would not be separated at the design test condition. It was not intended to test the model at high angles of attack where extensive separation would be present. A wing semispan of 0.90 m (which is about one half of the facility test-section width) was selected as a suitable size, giving a test-section blockage ratio of 1.3% at zero angle of attack. This is considered to be a reasonable value to avoid severe tunnel-wall lift-interference effects. The wing was constructed from 17-4 PH stainless steel to minimize dynamic-load deflections and corrosion. The measured construction tolerance was ±0.12 mm (0.005 in.) over most of the surface and ±0.24 mm (0.010 in.) at the extremities.

INSTRUMENTATION AND ACCURACY

The pressure instrumentation consisted of 229 orifices on the wing, installed at five spanwise stations (n = 0.1, 0.3, 0.5, 0.7, and 0.9), and 203 orifices on the tunnel-wall turntable. Orifice locations are listed in the sample tabulation of pressure data in table 2. In order to provide a smooth orifice installation, the wing orifices (0.50 mm diameter) were installed by the electronic-discharge method, in which an accurately controlled hole was burned perpendicular to (parallel to and below) the surface into a subsurface cavity. A tube-sized cavity (1.0 mm diameter) was burned parallel to and below the wing surface, a tube was inserted into the cavity, and the junction was sealed with epoxy. The tubes were installed in channels in both wing surfaces that were machined to within about 1.2 cm of each orifice (fig. 8(c)). The machined channel in the upper surface was filled with an epoxy resin and the channel on the lower surface was covered with a removable plate in order to provide access to the tubes and instrumentation wires. Finally, the wing surface was finished to its final dimensions. An accelerometer was installed in the wing tip to measure the frequency and amplitude of the vibrations of the steel wing which was designed to be rigid.

Surface static pressures on the wing and wall were measured using electronically actuated pressure-scanning valves containing pressure transducers that were connected to an automatic data recording system. Each survey of the wing and wall pressures required about 4 min to complete. The self-calibrating feature of the scanning valves provided an accuracy of about one-quarter percent of full scale of the ±8.62 N/sq cm (±12.5 psi) transducers, between ±0.006 and ±0.01 in pressure coefficient at transonic speeds. Tunnel test conditions were measured with precision manometer followers having an accuracy of about ±34.5 N/sq m, giving a Mach number accuracy of about ±0.002. Mach-number steadiness and controllability was about ±0.003 at M = 0.85 to 0.95. Tunnel-static pressure was measured on the tunnel wall 2.4 wing-root-chord lengths ahead of the wing-root leading edge. Angle of attack was set manually by rotating the wall turntable and setting the angle with an inclinometer with an accuracy of ~±0.03°.

TEST CONDITIONS AND PROCEDURES

Pressures were measured over the wing and wall at Mach numbers from 0.25 to 0.96, and Reynolds numbers from 3.4×10^6 to 10×10^6. Since the angle of attack could not be set by remote control, the investigation was conducted at the design angle of attack of 5°. The test conditions are listed in table 3.

Wing and wall pressures were measured without boundary-layer trips on the wing. Tufts and oil dots were placed at the wing-root junction to observe the flow. Next, boundary-layer trips were installed on the wing using sifted glass spherules at 4.5% chord and sublimation flow-visualization tests were made to determine an effective size. A supersaturated solution of biphenyl chemical ($C_5H_5C_6H_5$)

dissolved in petroleum either was sprayed on the model, which was then lightly sanded with smooth paper. Two final trip sizes were selected: 0.16 mm (0.0063 in.) diameter (No. 100 mesh) trips were used on the lower surface and outboard of 60% span on the upper surface; 0.23 mm (0.0090 in.) diameter (No. 70 mesh) trips were required on the upper surface over the inboard 60% span due to the larger leading-edge radius.

Next, wing oil-flow tests were made at several Mach numbers and Reynolds numbers. It was found that fluorescent oil on the metal surface could be adequately photographed in black light. The orifices were covered with clear, thin mending tape and the oil mixutre was applied in 1 cm wide spanwise stripes every 20% chord. These oil stripes flowed into a fairly uniform formation of chordwise streaks that were photographed during the test with a 70-mm camera mounted in the test-section ceiling plenum chamber. Oil streaks on the lower surface were observed and photographed after the test run. Finally, wing and wall pressures were measured with the wing boundary-layer trips. Additional pressures were measured at Mach numbers from 0.5 to 0.82 with the floor and ceiling slots taped to simulate solid walls.

Prior to the test in the 6-Ft Tunnel, the effect of wall mounting on the wing-root flow was investigated in a preliminary experikment in the Ames 2- by 2-Foot Transonic Wind Tunnel at Mach numbers up to 0.94, angles of attack up to 9° and a Reynolds number of 2.6 M using the Lockheed small-scale, 0.26 m semispan Wing C model.

DATA REDUCTION

Static-pressure measurements were reduced to standard pressure coefficients using the tunnel conditions measured at the beginning of each data set, following which the tunnel pressures were adjusted for total-temperature changes as required during the test run. Two or three data sets were recorded and the pressures were averaged because of noticeable effects of unsteady pressures. Pressure coefficients for each spanwise station were numerically integrated by Simpson's rule to determine wing-section normal-force and pitching-moment coefficients. Total normal-force and pitching-moment coefficients were also determined by Simpson's-rule numerical integration of the span-load and pitching-moment distributions. Machine plots of chordwise pressure distributions were also obtained and plots could be generated immediately after each test run.

TABULATED RESULTS

The wing and wall pressure coefficients and the integrated normal-force and pitching-moment coefficients are tabulated on microfiche records along with plots of chordwise pressure distributions. These are provided in a pocket in the back of

this report (appendix A) for the test conditions listed in table 3. A sample tabulation is given in table 2. All symbols are defined in the Nomenclature section.

DISCUSSION

Wing-Wall Junction Flow

The first research task was to determine the flow condition at the junction of the wing with the tunnel wall to determine if there was a problem with major flow separation. This problem was first investigated experimentally prior to the 6-Ft Tunnel test in the Ames 2- by 2-Foot Transonic Wind Tunnel on the small-scale Wing C model (0.26 m semispan). Oil flow tests were made at Mach numbers up to 0.95 and angles of attack up to 7° at Re = 2.6 M. The tunnel-wall boundary layer, calculated to be about 1.9 cm, is about the same ratio of the wing semispan as that for the 6-Ft Tunnel where the boundary-layer thickness has been measured to be about 8 cm. The oil streaks indicated that the flow was not separated at the wing root; however, there was a slight outflow over the rear third of the root.

In the 6-Ft Tunnel the wing-root flow was observed at $a = 5°$ using oil dots and tufts, and the flow was similar to the flow observed in the 2-Ft Tunnel; no wing-root flow separation was observed.

Boundary-Layer Trips

The next research task was to determine the required size of the boundary-layer trips. A sublimation test was first made with no boundary-layer trips at Re = 10 M. A photograph taken by a camera in the tunnel ceiling plenum chamber, after a long run of about 30 min is shown in figure 9(a). Sublimation occurred back to about 10% chord, indicating that the flow is already turbulent in this region. Unexpectedly, over the rest of the wing the biphenyl was only partially sublimed, even though the boundary layer was certain to be turbulent over most of the wing. It is felt that the large size of the wing and the resulting large boundary-layer thickness inhibited the sublimation process.

Since some tests were to be conducted at lower Reynolds numbers, it was decided not to depend on natural transition and sublimation tests were made with boundary-layer trips placed at 4.5% chord, which was the same location at which trips were placed on the small-scale model (refs. 1 to 4). The photograph (fig. 9(b)) of the final sublimation test with the final size of glass beads, selected to ensure transition at the trip location (see Test Conditions for size), shows that the biphenyl sublimed immediately behind the trips due to forced transition.

Oil-Flow Visualization

Figure 10 shows photographs of oil-flow tests at $M = 0.85$ and 0.82 at $Re = 10$ M. From previous experiments and calculations, the oil streaks over the wing surface are known to represent the surface skin friction lines. Tests were first made at the design Mach number of 0.85 (fig. 10(a)). The most prominent features in the flow pattern were the 3-D flow separation that occurred in the outer 30% of the semispan and the lack of 3-D boundary-layer flow over the rest of the wing. A faint trace of the main wing shock wave can also be seen by the slight S-curvature in the oil streaks, crossing the inboard 2/3 semispan, between 15% and 25% chord.

The local separation in the outer third of the semispan was caused by a strong shock-wave/boundary-layer interaction, as determined from the measured pressure distributions. A separation front exists along the shock wave at about 40% chord with flow around each end forming vortices, indicated by the focii at each end of the oil-flow pattern. The flow separation is localized to the outboard region, but it is not part of the wing-tip vortex flow. Also, the oil streaks show that the flow is not separated at the trailing edge. The pair of vortices from the pair of focii must lift off the wing and trail downstream. Air flows around and under the vortices coming together between the vortices; the air entrained in the vortex circulation flows forward and exterior air flows rearward, forming a saddle point on the surface, which can be seen in the oil flow.

The unexpected flow separation at the design test condition has inspired some attempts to calculate the flow pattern using the full Navier-Stokes computations. One such attempt is reported by Monsour (ref. 10). The results of this computation do not appear to match the results of the present oil-flow pattern. However, it is very interesting that Monsours's computations are similar to the oil-flow pattern obtained on the small-scale model in the Ames 2-Ft Tunnel at $a = 8°$ and 9° (not shown), in which a leading-edge separation vortex was prominent.

Inboard of the separated area the surface-flow angles were small over most of the wing, less than 10°, except near the leading edge. At the trailing edge the measured flow-direction angle was 8° outboard at the midsection. The predicted inviscid surface-flow-direction angle at the trailing edge (fig. 5(a) and (b)) is about 5° inboard so that the total change in flow angle through the boundary layer was only about 13°; hence, the boundary-layer flow is not very three dimensional.

Next, the Mach number was reduced to 0.82 (fig. 10(b)) where the flow separation disappeared. Only the weak design shock wave is observed in the oil-flow pattern, indicated by a slight S-shape in the oil streaks near 20% chord; this shock wave did not separate the flow. The most prominent feature is the lack of three-dimensionality in the flow pattern. The flow-direction angles were less than 10°, except near the leading edge. This was also true of the flow on the lower surface, as determined by post-test visual observation. Early in the design, it was expected that a low-aspect-ratio wing with large leading-edge sweep angle would have a large significant 3-D boundary-layer flow. Evidently, this is not necessarily the case. These results and those that follow show that this lack of three dimensionality

results from the wing design process in which the wing was optimized for a mild shock wave and a mild pressure recovery.

At this high Reynolds number of 10×10^6, the available Mach number range was limited, and so the Reynolds number was reduced to 6.8×10^6 and the effect of Mach number on the oil-flow patterns was investigated at $M = 0.70$, 0.82, 0.85, 0.90,. and 0.95 (fig. 11). This change in Reynolds number produced no effect on the oil-flow patterns at $M = 0.82$ and 0.85 (figs. 11(b) and (c)). At $M = 0.70$ the oil-flow pattern is similar to the attached-flow pattern at $M = 0.82$, except for the absence of the shock wave. Increasing the Mach number to 0.90 and 0.95 the flow separation that existed at $M =$, 0.85 over the outer 30% of the wing increased in extent and moved slightly rearward. Inboard of the separated region the flow-direction angles were still not very large at $M = 0.90$; however, at $M = 0.95$ a large outboard flow developed near the trailing edge where the boundary layer must have been highly three dimensional.

Decreasing Reynolds number to 3.4×10^6 produced no change in the oil-flow patterns at $M = 0.82$ and 0.85 (fig. 12).

Vapor-Trail Flow Visualization

In order to visualize the vortex flow off the wing at $M = 0.85$ to 0.95 where separation occurred in the oil-flow patterns, the water-vapor content of the tunnel was increased until the vortex trails could be observed with the ceiling lights on. The trails could be seen only vaguely and could not be photographed; however, it could be seen that the separated flow field was unsteady, oscillating irregularly at a low frequency. In this type of flow dynamics the possible contributing influence of the wind-tunnel flow dynamics to the model flow dynamics is unknown.

Shock-Wave-Induced Unsteady Pressures

The output of the pressure transducers was recorded on an oscillograph to check for both pressure lag and unsteadiness. It was found that the lag was small for the 0.51 mm orifices with approximately 2.5 m of 0.8 mm I.D. tubing; however, a noticeable unsteady pressure existed for the midsemispan for $M = 0.82$ as seen in the oscillograph traces in figure 13. The maximum unsteadiness existed at the shock-wave location at about 15% chord. The largest pressure fluctuations occurred at nearly regular intervals of about 3 Hz. Ahead of the shock-wave locations the pressures were relatively steady; however, behind the shock wave the pressures were unsteady to the trailing edge. In order to obtain mean values of pressure, two sets of data were recorded and averaged to partially compensate for the pressure unsteadiness. Figure 14 shows a comparison of one, two, and three-cycle averaged data with five-cycle averaged data. The shock-wave position changed as much as 2% chord from the averaged position, and the pressure coefficients behind the shock wave changed as much as 0.05 from the averaged value for a distance of about 20% chord. Shock-wave-induced unsteady pressures were recorded over the Mach number range of 0.80 to

0.95 in which the unsteady region moved rearward with the rearward movement of the shock-wave system.

Wing-Tip Accelerometer Measurements

The wing was constructed of steel to minimize the effects of flow dynamics on model dynamic response. Accordingly, the dynamic characteristics of the steel wing were investigated at $M = 0.80$ to 0.95 using the accelerometer mounted in the wing tip, oriented in the vertical direction. The maximum calculated wing-tip deflection due to flow dynamics was about 0.08 mm as determined from the accelerometer measurements at $M = 0.80$ TO 0.95.

Comparison of Measured and Computed Wing Pressures

Comparing large-scale and small-scale wing data with each other and with predictions can be difficult because of wall interference and model boundary-layer effects. The lift interference induced by the wind-tunnel wall can be different for each experiment. The effective thickness of the model boundary layer can increase the effective thickness of the wing, decambering the highly cambered aft wing section of supercritical airfoils and significantly affecting the predictions of lift and pitching moment. Therefore, it is interesting to compare the results from the two Wing C experiments with those from two wind tunnels with predictions. These comparisons supplement the comprehensive analysis of the small-scale data by Hinson and Burdges (ref. 3), using three configurations, Wings A, B, and C, high, medium, and low aspect-ratio wings.

The discussion considers the questions of the cause of the local-flow separation at the design test condition of $M = 0.85$, the general success of the predictions at several Mach numbers, the correlation of the two experiments, and the wall effects in the two tunnels, including the effect of taping the suction slots to simulate solid walls.

Figures 15 present experimental chordwise pressure distributions for the maximum test Reynolds number of 10×10^6 at two Mach numbers: $M = 0.82$ for unseparated flow, and $M = 0.85$, the design Mach number. Other measured pressures are presented in appendix A.

Before discussing the pressure distributions it is useful to understand the development of the shock-wave pattern with increasing Mach number (this is shown in fig. 16, taken from the small-scale results (ref. 3)). Note that the design shock wave occurs first nearly parallel to the leading edge. With increasing Mach number, a second shock wave forms that is nearly perpendicular to the wing root. The two shock waves coalesce in the planform to form a lamda shape, which is a well known transonic shock-wave pattern. For higher aspect ratio wings, the two shock waves merge into a single shock that extends over the outboard panel (ref. 3).

Figures 17 to 19 show a comparison of the experimental and predicted chordwise-pressure distributions and the corresponding spanwise-load distributions at the design Mach number of 0.85. This is an especially interesting case, since it has been shown by oil-flow visualization that local-flow separation occurred near the wing tip of the large-scale model. Results from the two Wing C experiments are shown: the present large-scale wing test and the previous small-scale wing test of Lockheed Georgia, both for $Re = 10 \times 10^6$ (based on mean aerodynamic chord). Predictions from two transonic wing codes are also shown.

FLO22 and TWING codes- Predictions from two 3-D, full-potential-flow transonic wing codes are included in the comparison of pressure distributions in figure 17. FLO22 is the nonconservative code used to design the wing (see Wing Design and figs. 3 to 7); TWING is a conservative code by Holst (refs. 11, 12, and 13) developed since the analysis by Hinson and Burdges (ref. 3). The TWING code has been shown to be successful for a wide range of wing shapes, from transport to fighter types (ref. 13). The code is included in figure 17 because it was used extensively in the conduct and present test and the analysis of the results because of its efficient algorithm and subsequent short computation time.

The predictions from FLO22 and TWING codes are generally in good agreement at the design Mach number of 0.85. At the midspan station TWING code shows a hump in the pressure distribution behind the design shock wave near the midchord, owing to the development of the second shock wave (fig. 16). It is typical of conservative codes that they capture the shock waves better than nonconservative codes. Consequently, the TWING results predict a slightly higher effective Mach number. This is seen further in the discussion of figures 20 and 21 which shows pressure measurements and TWING computations at several Mach numbers. At $n = 0.9$ (fig. 17) the location of the shock wave is clearly more rearward for TWING code, and both methods show a stronger than desired shock wave (fig. 3).

Small-scale Wing C- The first wing-pressure tests of the Wing C configuration were conducted by Lockheed Georgia using the small-scale semispan model in their high Reynolds-number facility at $Re = 10 \times 10^6$ (refs. 1-3). Those results are discussed first.

In a creditable analysis Hinson and Burdges (ref. 3) compared the small-scale-wing pressures from Wings A, B, and C with the predictions of several inviscid computer codes, including the design code FLO22 and a version of FLO22 modified to include corrections for boundary-layer displacement-thickness effect. (They did not include TWING code which did not exist at that time.) For comparisons with computations, the method of matching leading-edge pressures was used to select an experimental angle of attack of 5.9°, for which the experimental and predicted (FLO22 code, $a = 5°$) leading-edge pressures agree. This artifact cannot be used with all codes, but the FLO22 and TWING codes are noted for reasonable predictions of leading-edge pressures over a large range of sweep angles. The analysis assessed the tunnel wall lift- and blockage-interference effects by using measured floor and ceiling pressures, and the effective angle of attack as boundary conditions in the computations. It was found that the experimental wall pressures at the effective

angle of attack of 5.9° agreed closely with the computed free-air pressures at $a = 5°$, thus giving more credence to the comparison of computation and experiment.

The small-scale Wing C pressures are shown in figure 17 for $M = 0.85$ and $a = 5.9°$ (as selected in ref. 3). The pressures agree rather well with the computations; however, Hinson and Burdges concluded that the correlations were not quite as good for Wing C as for Wings A and B (ref. 3). The inclusion of viscous effects in the FL022 computation made little difference in the correlation for Wing C; however, significant improvements were reported for Wings A and B. The fact that it was necessary to use a higher experimental angle of attack (5.9°) than the design angle (5°) indicates that the tunnel-wall porosity was more than adequate to compensate for the wall-lift-interference effect, which tends to increase the lift coefficient at a given angle of attack.

The span-load distribution in figure 19 shows that at $a = 5.9°$ where the leading-edge pressures generally agree, the experimental loading is slightly higher across the span than that given by FL022, and $CN = 0.54$, compared to 0.52 for FL022.

Large-scale Wing C- For the present large-scale wing data, plotted in figures 17 and 18 for $M = 0.85$ and $a = 5°$ and $Re = 10 \times 10^6$, the frontal-area blockage of 1.3% chord is about the same as for the small-scale wing; however, the slotted-tunnel floor and ceiling pressures were not measured, so that the effect of lift interference could not be estimated as readily as for the small-scale wing (ref. 3). Fortunately, the lift interference must have been small, because the large-scale leading-edge pressures (figs. 17 and 18) just happen to agree with the predicted leading-edge pressures over the semispan for the design angle of attack of 5° and also with the small-scale leading-edge pressures at the selected angle of attack of 5.9°. Consequently, the large-scale pressures can be compared with the design pressures ($a = 5°$) without having to recompute the prediction at another angle of attack, and also, with the small-scale pressures without having to select another experimental angle of attack.

Inboard and midspan pressures. At the inboard and midspan stations ($n = 0.1$ and 0.5, figs. 17(a) and (b)), the upper-surface peak pressures decrease noticeably from the inboard to the mid section as predicted (fig. 4). The peak local Mach numbers near the leading edge, listed in figure 17, increase from 1.18 at $n = 0.1$ to as high as 1.56 at $n = 0.5$, for which the normal Mach number is 1.1 for a sweep angle of 45°. Thus, the design is successful in avoiding excessive local normal Mach numbers (higher than 1.2). The recompression occurs without a noticeable shock wave at the inboard section, in agreement with predictions, but at the midsection near $x/c = 0.17$ a mild shock wave forms that is oblique (supersonic local Mach number behind the shock wave). The shock wave is nearly parallel to the leading edge (according to the oil-flow photographs, figs. 9 to 11) and agrees approximately with the predicted location (figs. 4, 17, and 18).

The pressures near the trailing edge on both surfaces are more negative than predicted. This indicates a possible decambering effect, caused by the effect of the boundary-layer displacement thickness on the effective geometric thickness and

camber. This effect was smaller for the small-scale Wing C data, but it was strong for Wings A and B (ref. 3, discussed later in Other Existing Wing Data of Relevant Interest) at the Reynolds number of the tests (Re = 10×10^6).

The large-scale pressure distribution has a slight bump behind the shock wave, which is the beginning of the formation of the second shock wave, thus indicating a higher effective wind-tunnel Mach number than the small-scale data. This bump increases as the Mach number increases (figs. 16, 20, and 21, n = 0.5). The large-scale results agree more closely with the TWING code, which captures the second shock wave, than with the FL022 code. The effective Mach number difference between the two codes would appear to be about 0.03, according to the experimental results in figure 20 and the computations in figure 21. There is about the same effective Mach number difference between the two experiments.

The question of the "correct" Mach number can be resolved with wall-boundary measurements. Wall measurements were not obtained with the large-scale wing; however, they were obtained with the small-scale wing (ref. 3), and these results can be used to deduce a "correct" Mach number. Actually, two measurements are required for a complete analysis: wall-pressure distributions and either wall normal-velocity or local-flow-angle distributions (ref. 14). It is also important that side-wall pressures and normal velocities be included in the global computations. For the small-scale wing tests, the upper- and lower-surface near-wall pressures were measured with pressure rails at three rows and one side wall row, but no vertical velocities were measured. To cover the latter effect, it was assumed that the integral of the flow angle distribution could be represented by the effective angle of attack, determined by the separate process of matching leading-edge pressures. Using this effective angle (in this case, a = 5.9°) and the measured near-wall pressures, computations were made that indicated that the small-scale experimental Mach number was about 0.005 higher than the computed value (ref. 3). If this analysis is accepted as correct, the small-scale pressure distribution in figure 17(b) should be closest to wall-interference-free data. Hence, the FL022 code was reported in reference 3. To be more nearly correct than the conservative code (FL027 in ref. 3, comparable to TWING in fig. 17). Thus, the conservative codes slightly overestimate the strength of the shock waves, according to reference 3, which is well known from previous computational/experimental comparisons.

Outboard pressures. At the outboard station (n = 0.90, fig. 17(c)) a slightly higher effective Mach number is also evident in the large-scale data by the more rearward location of the shock wave than for the small-scale data. Except for shock-wave position, the predictions agree rather well with each other and with the experiment (considering that this station is close to the wing tip). Both predictions and experiment show a stronger shock than desired (fig. 3) because of a supersonic expansion behind the leading edge. Near the leading edge the local Mach numbers are as high as 1.54, about the same as at the midspan station n = 0.5, but occurring farther rearward at x/c ~ 0.35. The normal Mach number for a sweep angle of 45° is 1.09; however, the effective sweep at x/c = 0.35 could be less, say 40° or less, raising the local effective normal Mach number beyond 1.2 for local separation. The average shock-separation-front angle appears to be less than 40°. The

shock-wave/boundary-layer interaction causes the three-dimensional flow separation, indicated by the oil-flow results (fig. 10(a)). The pressure coefficient at the trailing edge is -0.04, which indicates that the flow separation exists to the trailing edge. It is interesting that the pair of vortices rotate in an opposite sense to the tip vortices of the lifting wing (like a wing element at negative angle of attack); hence, the separated cell causes a local lift decrement, as expected.

Span load distribution- The span-load distribution in figure 19 for $M = 0.85$ shows that at $a = 5°$, where the leading-edge pressures generally agree, the experimental loading is higher across the span than that given by FL022 code, and $CN = 0.54$, compared to 0.52 for FL022 code. Since the large-scale wing is at the design angle of attack, the slightly higher loading indicates a small wall-interference effect. A slightly higher loading can be seen in the experimental pressure distributions (fig. 17), possibly owing to a higher effective Mach number according to the evidence given in the previous discussion.

Effect of Reynolds number- Pressure distributions and oil-flow patterns were obtained at other Reynolds numbers from 3.4×10^6 to 10×10^6 at $M = 0.85$. The results showed no significant effect of Reynolds number.

Effect of Mach number- Figure 20 shows the effect of Mach number from $M = 0.80$ to 0.95 on the experimental pressure distributions at the midsemispan station, $n = 0.50$. The leading-edge pressures decrease as expected and the first shock wave moves slightly rearward; however, the most noticeable effect is the appearance, growth, and size of the rearward movement of the second shock wave. These effects are well predicted by the TWING code, shown in figure 21, which demonstrates the great utility, because of the short run times, of this code when studying many off-design conditions. The second shock wave appears in the FL022 code at a higher Mach number (see fig. 23(f) for $M = 0.90$).

Predicted effects of angle of attack- The usefulness of the TWING code in analyzing the possible effects of angle of attack on the experimental data dan be seen in figure 22 for $n = 0.50$ and $a = 1°$ to $6°$. These computations, available before the test, were used to consider the possible reduction in angle of attack to reduce the strength of the shock wave. A reduction of two or three degrees would be required, consequently the test angle of 5° was retained.

Off-design pressure distributions- Carpet plots of predicted wing pressures from FL022 code are presented in figure 23 for $M = 0.25$, 0.50, 0.70, 0.82, and 0.90 for $a = 5°$; also included is $M = 0.85$, $a = 7°$ (which is used later in the discussion of the effect of wall suction slots). At $M = 0.82$, the pressure distributions appear to achieve the design goals of a mild shock wave and a mild pressure recovery with no indication of any possible problems of flow separation. Stable solutions were obtained at all of these conditions for which strong shock wave appear in the solutions, even at the highest Mach number of 0.90 and the highest angle of attack of 7°.

Figure 24 presents a comparison between the predicted pressures by FL022 code and the experimental pressures at Mach numbers of 0.25, 0.50, and 0.82. The results

show excellent agreement at $M = 0.25$ and 0.50 and good agreement at $M = 0.82$. Note that, again, the leading edge pressures just happen to agree with the computations so that no adjustment in angle of attack was required in the computations. At $M = 0.82$ the experimental pressure distributions achieved the design goals (hoped for at $M = 0.85$) of a mild shock wave and a mild pressure recovery with no indication of any possible problems of flow separation.

Effect of wall suction slots- Figure 25 shows a comparison of experimental pressure distributions with floor and ceiling suction slots taped closed to simulate solid walls with those for slots open, with a porosity of 6% of the floor and ceiling area, for $M = 0.82$ and $Re = 6.8$ M. A large increase in lift blockage can be seen, increasing the normal-force coefficient from 0.52 to 0.65, which would correspond to an effective angle of attack of about 7° ($CN = 0.62$) according to the Lockheed small-scale tests (ref. 1). In addition, the position of the shock wave at the midspan station, $n = 0.5$, is about $x/c = 0.65$ with slots taped, which indicates that the effective Mach number is about 0.87, according to figure 20.

In order to further investigate these observations, a computation was made with FL022 code for $a = 7°$ and $M = 0.85$ (to compare with $a = 5°$ at $M = 0.85$). The carpet plot is presented in figure 23(e) and the results are compared with the experiment in figure 26. (For comparison, the carpet plot for $M = 0.85$ and $a = 5°$ is in figure 4.) The computations for free air at $M = 0.85$ and $a = 7°$ are similar to the experimental results with slots taped for $M = 0.82$ and $a = 5°$. The lift coefficient is about the same ($CL = 0.65$) and a second shock wave is prominent; however, the computed shock-wave position is $x/c \approx 0.45$ compared with the measured position of about 0.65. The more rearward position of the measured shock wave indicates that the effective Mach number with slots taped was higher than the computed Mach number of 0.85, as concluded earlier. Thus, a strong lift blockage ($\Delta a \sim 2°$) and drag (Mach number) blockage ($\Delta M \sim 0.5$) are indicated by the data with suction slots taped. Some flow separation is indicated at the trailing edge by the negative pressure coefficients.

It was intended to compare these experimental results to predictions by the FL029 computer code, a 3-D, full-potential, conservative, transonic wing code developed for computing the flow for a wing in a wing tunnel. However, experience at Ames (ref. 15) with various versions of the FL029 code have shown anomalies which indicate that the code is not performing adequately at this time for the in-tunnel case.

Retrospection of Wing C Design and the Problem of Local-Flow Separation

In retrospect, the potential for the occurrence of the shock/boundary-layer flow separation can be perceived by reexamining the predicted pressure distributions in figures 4(a) and (b). Recall that it was pointed out in the discussion of figure 17 that at $n = 0.90$ a stronger-than-desired shock wave was observed in both the predicted and the measured pressures. Now note in figure 4 that the desired design pressure distribution occurs outboard of $n = 0.90$ at about $n = 0.93$, and

that inboard of $n = 0.93$ a strong shock wave is predicted to occur at about $n = 0.63$. In addition, at $n = 0.78$, 0.83, and 0.88 the predicted pressures show a short supersonic expansion behind the leading edge that strengthens the shock wave and increases the possibility of separating the boundary layer. This is also seen in the measured pressure distributions at $n = 0.90$. Thus, it is possible that the flow separation could have been avoided in the design by eliminating this local supersonic expansion. On the other hand, this shock-wave/boundary-layer separation might still occur at Mach numbers above the design condition.

In a private discussion of this separation problem. Hicks (Ames Research Center) suggested that additional wing twist and camber might have achieved the desired results. However, it might also have been necessary to eliminate the requirement of linear lofting between the root and tip stations, imposed on the design to simplify the machining of Wing C. Thus, more than two design control stations should probably have been used for good transonic wing design. Further, it has been noted that when FL022 is used in transonic wing design, it is best to design for shockless flow at some Mach-number increment higher than the desired design value to avoid shock-induced separation. The present Wing C results indicate that this Mach number increment should be about 0.03, since the flow is unseparated at $M = 0.82$, but separated at the design Mach number of 0.85. In addition, it would be useful to supplement the FL022 prediction with a computation from a conservative code like TWING which captures the shock waves more clearly, but with some local effects on the pressures behind the shock. The time-efficient TWING code could then be used to compute many off-design conditions.

Other Existing Wing Data of Relevant Interest

The results from six other wing tests that are relevant to the results for Wing C are reviewed (figs. 1 and 27). These other results contribute to the present analysis in two ways: first, the conclusions from the computational/experimental analysis of the Wing C pressures cannot be generalized without including the investigation of other wing configurations, so that the analysis of small-scale Wings A and B (ref. 3) is a valuable contribution to the investigation and is reviewed below. Second, the lack of three-dimensional surface flow in the present Wing C oil-flow patterns at the design condition for unseparated flow introduces the question of what wing configurations might have significant 3-D surface flow at the design condition for unseparated flow. Evidently, a highly 3-D low-aspect-ratio wing configuration with large leading-edge sweep angle does not necessarily have a significant 3-D boundary-layer flow at the design condition. The investigations of the first five wings are cited to support this allegation, since the first five wings have the common characteristic that they had small 3-D surface flow at design conditions. The last wing (the transport wing) experienced strong 3-D boundary-layer flow and so those results are reviewed to determine what configuration differences caused the 3-D boundary layer.

Lockheed Georgia Wings A and B- As part of the comprehensive cooperative investigation by Lockheed Georgia, Hinson and Burdges (refs. 1-3 and fig. 1) designed and

tested two other small-scale wing models, designated A and B. In reference 3, the small-scale wing results were compared with the predictions of several inviscid computer codes. They reported that the correlations with FL022 code were surprisingly good for Wings A and B. It was especially noteworthy that when 2-D viscous effects were included in the FL022 computation, the prominent decambering effect owing to the boundary-layer displacement thickness effect on the aft pressure distributions were correctly predicted. The predicted decambering effect resulted in a significant reduction in predicted lift coefficient for Wings A and B. The final lift coefficients agreed with the predicted values for Wings A and B, indicating that the wind-tunnel wall effects were eliminated by the perforated ceiling and floor suction. This was not the case for Wing C results, for which the final lift coefficients were 0.54 (measured) and 0.52 (FL022, fig. 17). It is interesting that the effects of viscosity on the pressure distribution could be predicted so well for Wings A and B using the 2-D boundary-layer code.

Streett (ref. 16) investigated further the decambering effect of the boundary layer for Wing A using an existing 3-D compressible, integral boundary-layer method and concluded that a 3-D code would be necessary only near the wing tip for Wing A. No oil-flow results are available for Wings A and B. They were designed for weak shock waves and mild pressure recoveries, similar to Wing C, whose oil-flow results show almost 2-D surface-flow patterns for unseparated flow conditions.

NASA Dryden F-8 research airplane- A supercritical wing, similar in planform to Wing A, was flight tested on an F-8 research airplane at NASA Dryden Flight Research Center (fig. 27). Montoya and Banner show (fig. 19, ref. 17) that the measured boundary-layer flow angles were nearly zero at the trailing edge at angles of attack less than about 5° at Mach numbers up to about 0.90, which is similar to the results for Wing C. Oil-flow photographs from a wind tunnel model show nearly 2-D surface flow over most of the wing at $M = 0.90$ and $a = 3.5°$ (fig. 18, ref. 17).

FAA, Sweden, Saab 32 Lansen research airplane- Flygtekniska Forsoksanstalten (FAA), Sweden, made surface-pressure and boundary-layer measurements over the outer wing panel of a Saab 32 Lansen with an NACA 64A010 wing section, both in flight and in a wind tunnel. Bertelrud (ref. 18) reported that the curvature of the wall streamlines, deduced from oil-flow visualizations, was small over the main part of the wing.

NASA Ames Swept NACA 0012 semispan wing- Reference 15 describes an experimental investigation of the turbulent, subcritical, and supercritical flow over a swept, NACA 0012 semispan wing (fig. 27) in a solid-wall, high-Reynolds-number wind tunnel (Ames). Surface-pressure and laser-velocimeter flow-field measurements are presented and the results are compared with two inviscid wing codes. Although this wing was relatively thick and was not computer optimized, the oil-flow photographs show that the surface-flow pattern was not very three dimensional when the flow was unseparated. At $M = 0.82$ and 0.83 flow separation occurred in the outboard region of the upper surface, similar in appearance to that of Wing C. However, the separation occurs here because of the thick wing section (12% chord) which is less supportive of supercritical flow than the Wing-C sections and induces a strong shock

wave. Also, the model had no wing twist to alleviate the spanwise increase in effective angle of attack.

NASA Ames/McDonnell-Douglas transport wing- A cooperative experimental test program was conducted by Spaid of McDonnell Douglas (ref. 6) in the Ames 14-Foot Transonic Wind Tunnel using a transport wing configuration (fig. 27). Unlike the preceding models, this model was designed for a strong pressure-recovery gradient over the rear of the wing section to increase the aft wing volume. The design problem was to avoid separation from the strong adverse pressure gradient. Predicted inviscid wing pressures and surface streamlines by FL022 code are presented in figure 28 for the experimental conditions of $M = 0.825$ and $a = 4°$. The predicted pressures show the strong design shock waves and pressure-recovery gradients. The predicted inviscid, surface, streamlines are not very three dimensional; however, the streamlines have more curvature for the transport wing than for Wing C (fig. 5(b)).

A mini-tuft flow-visualization photograph from reference 5 is reproduced in figure 29, showing that the boundary-layer flow on the surface turned outboard over the last 15% chord. Flow angles at the trailing edge were measured as high as 30° outboard (reproduced in fig. 30). From these results it is evident that in some wing designs, aerodynamic optimization is relaxed to allow the boundary layer to approach separation, which induces significant 3-D boundary-layer flows. For these wings, test results should be analyzed to determine if the methods and accuracy of a boundary-layer are required.

CONCLUSIONS

The conclusions of the analysis of surface-pressure and oil-flow photographs from wind-tunnel tests of a large-scale semispan model of Ames/Lockheed Wing C are presented below. Wing C is a generic, transonic, supercritical, low-aspect-ratio configuration, designed for a Mach number of 0.85 and an angle of attack of 5°, using a 3-D transonic, potential-flow code (FL022) and an optimization routine. Pressures were measured at the design angle of attack over a Mach number range from 0.25 to 0.96 and a Reynolds number range of 3.4×10^6 to 10×10^6 with both the tunnel floor and ceiling suction slots open for most of the tests, and taped closed for some tests to simulate solid walls. A brief comparison was made with pressures measured in a small-scale model tested at the same Reynolds number and with predictions from two transonic wing codes: design code FL022 (nonconservative) and TWING code (conservative).

1. At the design Mach number and angle of attack of 0.85 and 5°, respectively, the oil-flow patterns showed that local-flow separation occurred in the outer 30% of the semispan, caused by a strong, local, shock-wave/boundary-layer interaction that was not a tip-vortex effect. The flow separation increased as the Mach number was increased to 0.95, but disappeared when the Mach number was reduced to 0.82. Though

undesired, this separation provides interesting data for calculations of viscous wing flows with shock-wave/boundary-layer separation.

2. At Mach 0.82 the flow was unseparated and the oil-flow pattern showed that the surface-flow pattern was not very three dimensional. The surface oil-streak angles were less than 10°, except near the leading edge. At the trailing edge, the flow angles were 8° outboard on the surface and about 5° inboard, outside of the boundary layer, according to inviscid computations, so that most of the boundary-layer flow is nearly two dimensional.

3. At Mach 0.82 the main wing shock wave was found to be unsteady at a low, irregular frequency of about 3 Hz, inducing unsteady pressures to the trailing edge; however, the model was inflexible and the model dynamic oscillations were negligible. The unsteadiness extended over a Mach number range of 0.80 to 0.95.

4. Comparing the large-scale and small-scale data with each other and with predictions are complicated by wall interference and model boundary-layer effects. For example, the normal-force coefficients at the design Mach number and angle of attack of 0.85 and 5° were 0.48 and 0.54 for the small-scale and large-scale models, compared with 0.52 for both the FL022 and TWING codes. Effective Mach number differences between the two models and the predictions could be as large as 0.03, according to the results. The small-scale study of Hinson and Burdges showed that a more definitive determination of the effective Mach could be made by using tunnel-wall measurements in the computations.

5. Matching leading-edge pressures (used by Hinson and Burdges in their analysis) appears to be one satisfactory method of contending with the difficulty of selecting an experimental angle of attack to correlate the experimental and predicted pressure distributions. Using this method, predictions by design code FL022 (nonconservative) generally agree well with the experiments to Mach numbers as low as 0.25, except for the details of the variations in shock position and aft loading. Comparisons with predictions by TWING code (conservative) at Mach 0.85 show generally good agreement with FL022 code and the experiments, except that TWING code captures the second shock wave more than FL022 code.

6. Wall interference effect was effectively demonstrated when the floor and ceiling suction slots were taped to simulate solid walls at Mach 0.82. The normal force coefficient increased tremendously from 0.52 to 0.65 (equivalent to about 7° in incidence) and the effective Mach number increase to about 0.87. It was intended to compare the pressures, measured with the suction slots taped, with predictions by the FL029 computer code, a 3-D, full-potential, conservative, transonic wing code developed for computing the flow for a wing in a wind tunnel. However, experience at Ames has shown anomalies that indicate that the code does not adequately predict the in-tunnel case.

7. In retrospect, the flow separation that existed at the design conditions might have been avoided by further iteration in the design because the inviscid pressure distributions indicate a slightly stronger shock wave than that desired in the region of the measured separation. Not surprisingly, it appears that more than

two defining stations are needed to design an efficient transonic wing. When FL022 is used in transonic wing design, one should design for shockless flow at a higher Mach number than the desired design value (an increment of about 0.03, according to the present oil-flow results).

8. Evidence from this study and from other cited wing studies indicates that wings that are optimized for mild shock waves and mild pressure-recover gradients generally have small 3-D boundary-layer flow (flow angles less than 10°) at design conditions for unseparated flwo. Further, for these wings, 2-D boundary-layer methods appear to be sufficient to predict the effects of boundary-layer thickness on the pressure distributions.

9. Evidence from another cited wing study indicates that in some wing designs optimization is relaxed to allow the boundary layer to approach separation, which induces significant 3-D boundary-layer flows near the trailing edge.

RECOMMENDATIONS

Based on the results of the Wing-C test program the following recommendations are offered.

1. That a transonic code for a wing in a tunnel with solid walls be developed to compare with the experimental pressure distributions with wall suction slots taped closed to simulate solid walls. A transonic, small-disturbance code exists for this problem; also, subsonic panel codes can treat the wing-tunnel case.

2. That the experimental pressure distributions at Mach numbers of 0.85 and above with local flow separation be used for comparison with Euler and Navier-Stokes codes.

3. That the cited 3-D boundary-layer results for the transport wing should be analyzed to determine (a) if three-dimensional boundary-layer methods for the case of unseparated flow are required, and (b) their accuracy.

APPENDIX

TABULATIONS AND PLOTS OF PRESSURE DATA

Microfiche records are enclosed on the inside back cover for the sets of tabulated data (1 fiche) and plots of chordwise pressure distributions (2 fiches) for the test conditions listed in table 3. A sample copy of one of the tabulations is shown in table 2.

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TABLE 1.- SECTION ORDINATES OF WING C AT ROOT AND TIP

N	X/C	Tip		Root	
		Z/C_U	Z/C_L	Z/C_U	Z/C_L
1	0.00000	0.00000	0.000000	0.00000	0.00000
2	.00241	.00730	-.006025	.00967	-.00503
3	.00961	.01542	-.009709	.01784	-.00941
4	.02153	.02261	-.012482	.02584	-.01244
5	.03806	.02830	-.015382	.03351	-.01480
6	.05904	.03285	-.018439	.04109	-.01696
7	.08427	.03653	-.020903	.04854	-.01863
8	.11349	.03928	-.022924	.05581	-.01995
9	.14645	.04115	-.024471	.06290	-.02089
10	.18280	.04221	-.025486	.06965	-.02130
11	.22221	.04261	-.026195	.07586	-.02142
12	.26430	.04253	-.026280	.08108	-.02101
13	.30866	.04202	-.025949	.08493	-.02023
14	.35486	.04109	-.025082	.08718	-.01884
15	.40245	.03982	-.023888	.08770	-.01704
16	.45099	.03812	-.022217	.08648	-.01462
17	.50000	.03613	-.020079	.08368	-.01172
18	.54901	.03384	-.017094	.07951	-.00798
19	.59755	.03135	-.013470	.07427	-.00362
20	.64514	.02864	-.009348	.06818	.00112
21	.69134	.02584	-.005664	.06142	.00518
22	.73570	.02298	-.002667	.05418	.00825
23	.77779	.02006	-.000695	.04682	.01003
24	.81720	.01710	.000481	.03956	.01050
25	.85355	.01415	.000802	.03256	.00972
26	.88651	.01124	.000588	.02605	.00807
27	.91573	.00855	.000108	.02016	.00589
28	.94096	.00618	-.000269	.01491	.00362
29	.96194	.00422	-.000561	.01049	.00142
30	.97847	.00272	-.000598	.00701	-.00028
31	.99039	.00172	-.000501	.00452	-.00141
32	.99759	.00110	-.000698	.00315	-.00233
33	1.00000	.00082	-.000821	.00270	-.00270

TABLE 2.- SAMPLE TABULATION OF DATA FROM APPENDIX

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RUN:SEQ
165:3

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.820	2.986	6.79	1505	968	538.1	455.4	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.440	0.083	0.523	-0.0206	-0.0131	-0.0338	0.2245	29.68	40.91	31.46	42.95	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.343	0.097	0.440	-0.0418	-0.0184	-0.0602	37.16	44.04	38.67	0.631	0.141
0.296	0.437	0.086	0.524	-0.0342	-0.0232	-0.0574	32.82	51.82	35.96	0.639	-0.002
0.500	0.506	0.093	0.599	-0.0282	-0.0277	-0.0559	30.57	54.94	34.34	0.598	-0.107
0.697	0.538	0.082	0.619	-0.0287	-0.0259	-0.0546	30.34	56.74	33.81	0.487	-0.137
0.894	0.550	0.022	0.572	-0.0462	-0.0155	-0.0616	33.39	96.53	35.78	0.328	-0.106

WING LOWER SURFACE COEFFICIENTS

2Y/B X/C	0.099	0.296	0.500	0.697	0.894
0	0.543	0.543	0.543	0.543	0.543
0.01			0.404		
0.02			0.329		
0.03			0.278		
0.04			0.244		
0.05	0.225	0.214	0.214	0.188	0.091
0.10	0.151	0.122	0.105	0.094	0.020
0.15	0.114	0.077	0.068	0.037	-0.024
0.20	0.096	0.043	0.041	0.023	-0.054
0.30	0.059	0.010	0.004	-0.008	-0.096
0.40	0.027	-0.015	-0.012	-0.022	-0.102
0.50	-0.002	-0.032	-0.010	-0.028	-0.087
0.55			******		
0.60	0.036	0.042	0.058	0.045	-0.007
0.65			0.110		
0.70	0.120	0.146	0.154	0.146	0.110
0.75	0.146	0.172	0.186	0.181	0.145
0.80	0.161	0.185	0.204	0.194	0.154
0.85	0.150	0.186	0.197	0.190	0.158
0.90	0.129	0.150	0.173	0.164	0.129
0.95	0.073	0.107	0.122	0.125	0.085
1.00	0.019	0.039	0.039	0.047	-0.017

WING UPPER SURFACE COEFFICIENTS

2Y/B X/C	0.099	0.296	0.500	0.697	0.894
0	0.449	0.357	0.273	0.219	0.237
0.003			-0.144		
0.006			-0.362	-0.392	
0.01	-0.275	-0.508	-0.619	-0.642	-0.554
0.02	-0.574	-0.870	-0.949	-0.933	-0.779
0.03	-0.657	-0.967	-1.113	-1.066	-0.859
0.04	-0.644	-1.107	-1.155	-1.059	-0.816
0.05	-0.642	-1.152	-1.239	-1.281	-1.010
0.06	-0.608	-1.110	-1.264	-1.229	-1.018
0.08	-0.587	-1.093	-1.234	-1.187	-0.986
0.10	-0.561	-0.981	-1.260	-1.200	-0.993
0.125	-0.526	-0.934	-1.247	-1.224	-1.001
0.15	-0.517	-0.766	-1.262	-1.241	-1.006
0.175	-0.514	-0.717	-0.859	-1.211	-1.016
0.20	-0.500	-0.665	-0.762	-1.198	-1.044
0.225	-0.489	-0.614	-0.712	-0.887	-1.021
0.25	-0.463	-0.574	-0.690	-0.720	-1.022
0.30	-0.445	-0.543	-0.631	-0.656	-0.909
0.35	-0.433	-0.526	-0.600	-0.643	-0.810
0.40	-0.439	-0.494	-0.576	-0.621	-0.722
0.45	-0.417	-0.456	-0.519	-0.589	-0.587
0.50	-0.397	-0.426	-0.479	-0.534	-0.463
0.55	-0.365	-0.385	-0.426	-0.461	-0.408
0.60	-0.333	-0.364	-0.370	-0.375	-0.371
0.65	-0.282	-0.332	-0.319	-0.318	-0.322
0.70	-0.255	-0.280	-0.283	-0.265	-0.277
0.75	-0.246	-0.228	-0.216	-0.213	-0.256
0.80	-0.195	-0.198	-0.184	-0.161	-0.219
0.85	-0.147	-0.137	-0.124	-0.114	-0.169
0.90	-0.086	-0.092	-0.055	-0.056	-0.130
0.95	-0.029	-0.006	-0.000	0.002	-0.082
1.00	0.019	0.039	0.039	0.047	-0.017

TABLE 2.- CONCLUDED.

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WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	-0.006				-0.397				
11.35	0.270	0.083				-0.395				
12.35	0.294	A				-0.388		-0.362		
14.35	0.342	0.067				-0.398				A
15.35	0.366	0.059				-0.412				-0.342
16.35	0.390	0.063				-0.404				0.286
17.35	0.413	0.049				-0.385		-0.362		-0.311
18.35	0.437	0.040				-0.392				-0.317
19.35	0.461	0.050				-0.381				-0.299
20.35	0.485	0.037				-0.380				-0.310
22.35	0.533	0.031				-0.350				-0.309
23.35	0.556	0.025				-0.363				-0.292
24.35	0.580	0.051				-0.324		-0.331		-0.303
25.35	0.604	0.054				-0.306				-0.289
26.35	0.628	0.066				-0.326				-0.290
27.35	0.652	0.074				-0.313				A
30.35	0.723									-0.230
31.35	0.747		0.139		-0.234		-0.234	-0.213	-0.231	-0.221
32.35	0.771		0.144		-0.219		-0.211		-0.209	-0.197
33.35	0.795		0.155		-0.201		-0.196		-0.180	-0.173
34.35	0.818		0.147		-0.181		-0.167		-0.168	-0.152
35.35	0.842		0.142		-0.150		-0.175		-0.156	-0.167
36.35	0.866		0.133		-0.139		-0.147		-0.139	-0.144
37.35	0.890		0.122		-0.104		-0.112		-0.134	-0.127
38.35	0.914		0.103		-0.073		-0.101		-0.105	-0.107
39.35	0.938		0.100	-0.034	-0.049		-0.075		-0.088	-0.094
40.35	0.961		0.070	-0.001	-0.042		-0.048		-0.075	-0.096
41.35	0.985		0.054	0.025	-0.002		-0.021		-0.059	-0.069
42.35	1.009		0.055	0.042	0.022		-0.015		-0.040	-0.149
44.85	1.069		0.027	0.005	0.021		-0.006		-0.042	-0.066
45.85	1.092		0.038	0.018	0.012					
46.35	1.116		0.038	0.023	0.012					

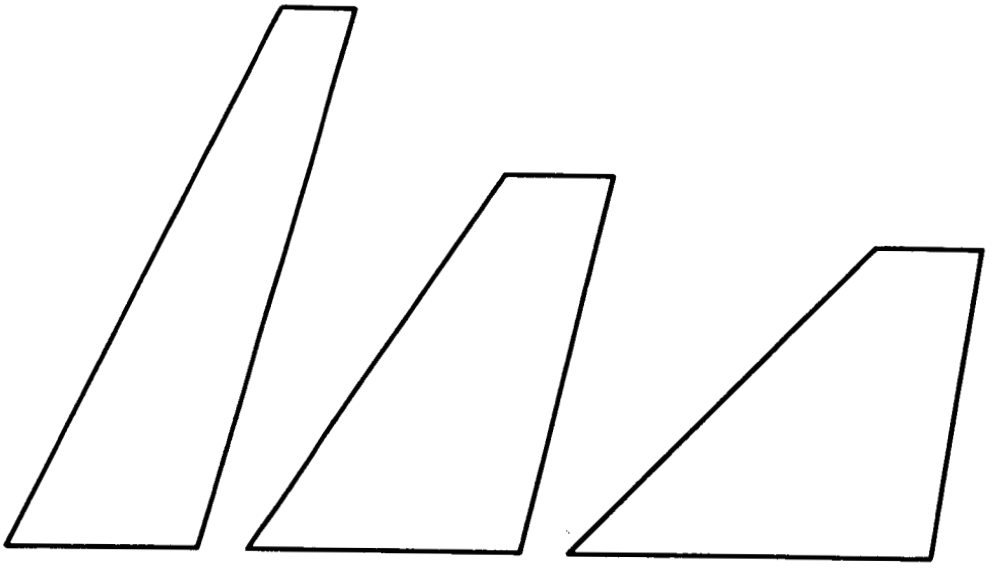
NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.205	-0.206	-0.149			
13.75	0.328		-0.276	-0.263	-0.214	-0.147			
10.75	0.256		-0.311	-0.303	-0.221	-0.144	-0.069	-0.149	
7.75	0.185	-0.362	-0.362	-0.331	-0.213	-0.137	-0.058		-0.035
6.75	0.161					-0.231	-0.139	-0.059	-0.040
5.75	0.137	-0.387	-0.390	-0.340	-0.236	-0.155	-0.035	-0.044	-0.013
4.75	0.113					-0.235	-0.155	-0.046	-0.038
4.25	0.101	-0.388	-0.385	-0.324					
3.75	0.089					-0.234	-0.147	-0.021	-0.015
2.75	0.066					-0.235	-0.133	-0.029	-0.008
1.75	0.042					-0.234	-0.026	-0.013	0.010
0.75	0.018					-0.234	-0.139	-0.002	0.022
-0.25	-0.006							0.025	0.042
-1.25	-0.030					0.139	0.133	0.054	0.055
-2.25	-0.054					0.140	0.134	0.066	0.067
-2.75	-0.066	A		0.049	0.051				
-3.25	-0.077					0.107	0.117	0.073	0.058
-4.25	-0.101	0.085	0.045	0.045	0.099	0.113	0.082	0.066	0.048
-5.25	-0.125					0.109	0.107	0.080	0.062
-6.25	-0.149	0.089	0.056	0.062	0.087	0.119	0.079		-0.075
-9.25	-0.220		0.065	0.052	0.075	0.081	0.083		
-12.25	-0.292		0.060	0.054	0.080	0.074			
-15.25	-0.363			0.057	0.077	0.069			

A)WTSAV020 OUT OF RANGE.

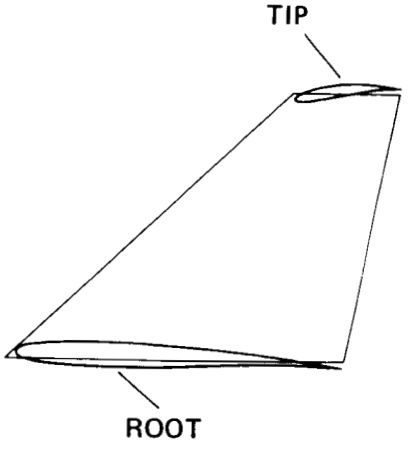
TABLE 3.- TEST CONDITIONS FOR MEASURED WING AND WALL PRESSURES

Reynolds number	Mach number	Run Number (listing)
3.4×10^6	0.25	206
3.4×10^6	.82	183
4.6×10^6	.82	182
4.6×10^6	.86	184
5.7×10^6	.82	181, 205
6.8×10^6	.50	204
	.60	203
	.70	202
	.74	201
	.78	200
	.80	197
	.81	196
	.82	165, 195
	.83	194
	.84	193
	.85	192, 199
	.86	191, 198
	.88	190
	.90	189
	.92	188
	.94	187
	.95	186
	.96	185
7.9×10^6	.82	180
9.1×10^6	.82	179
9.1×10^6	.86	178
10×10^6	.80	174
	.81	173
	.82	172
	.83	171
	.84	170
	.85	169
	.86	168
	.88	167
	.90	166



WING –	A	B	C
ASPECT RATIO –	8.0	3.8	2.6
L.E. SWEEP –	27°	35°	45°
TAPER RATIO –	0.4	0.4	0.3

Figure 1.- Sketch of small-scale wing models for Lockheed-Georgia tests in their high Reynolds number facility.



AR = 2.6 $\quad \lambda = 0.3$

$\Lambda_{L.E.} = 45°$ $\quad \Lambda_{T.E.} = 9.7°$

$S/2 = 0.625\ m^2$ $\quad b/2 = 0.902\ m$

$t_{ROOT}/C_R = 0.06$ $\quad t_{TIP}/C_T = 0.10$

$C_{ROOT} = 1.066\ m$ $\quad C_{TIP} = 0.318\ m$

M.A.C. = 0.760 m

$\alpha_{O,R} = 2.38°$ $\quad \alpha_{O,T} = -5.79°$

Figure 2.- Wing C geometry.

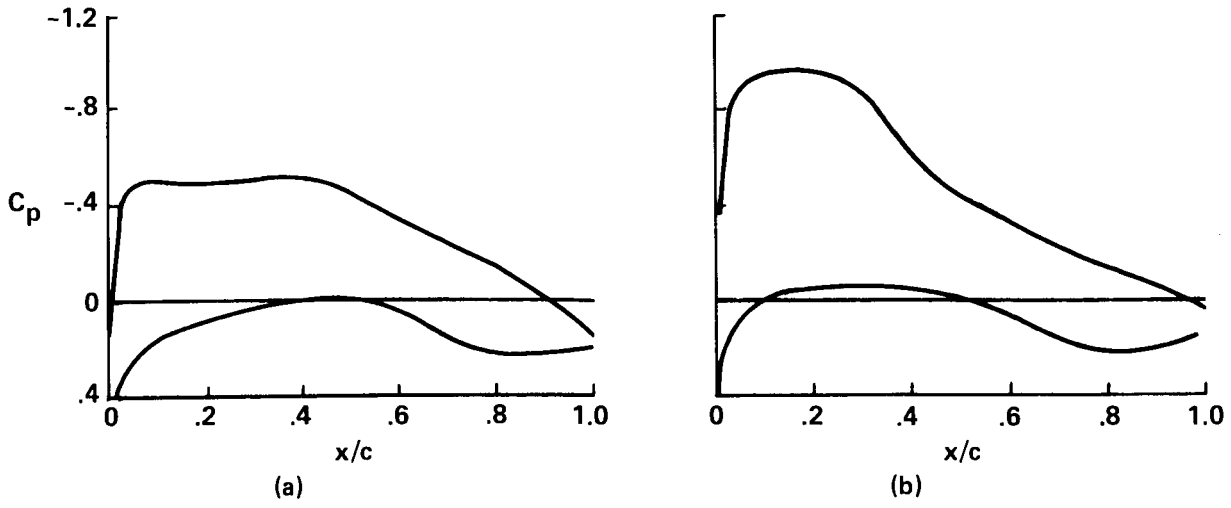
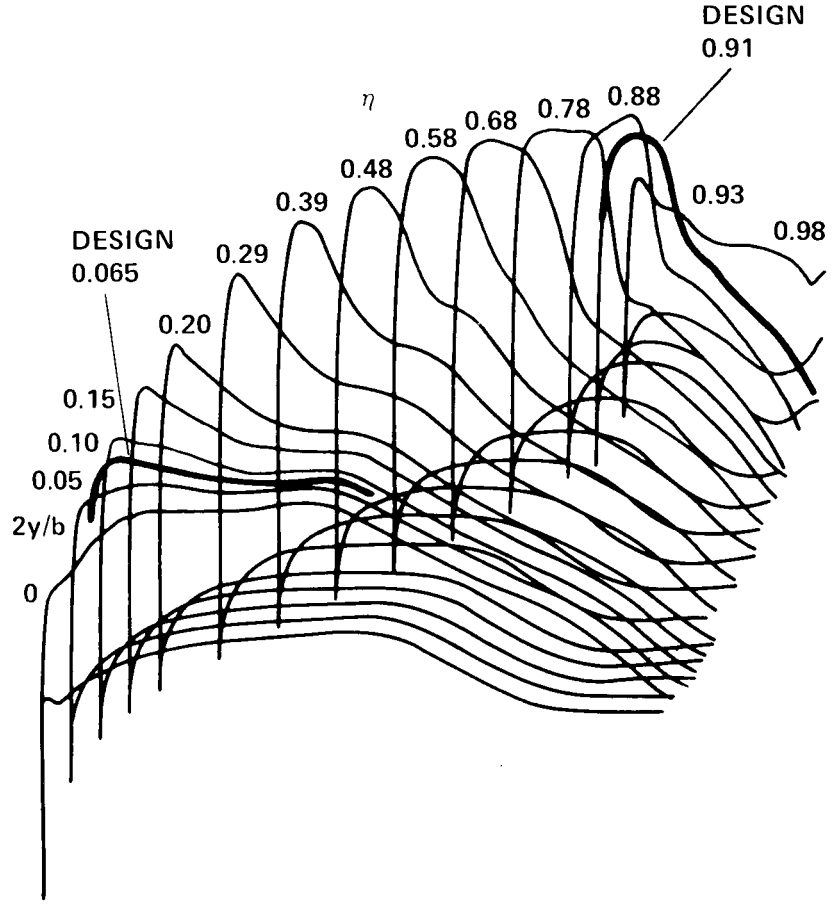
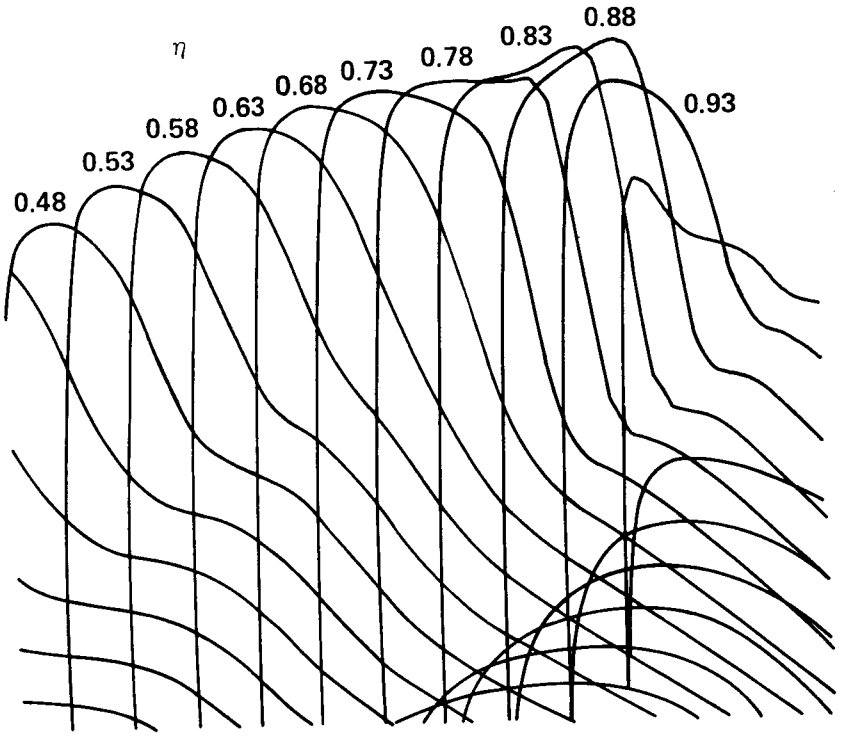


Figure 3.- Specified design chordwise-pressure distributions; M = 0.85, a = 5°.



(a) Carpet plot.

Figure 4.- Predicted inviscid wing chordwise pressure distributions by FL022 code at design conditions; M = 0.85, a = 5°.



(b) Selected distributions for n = 0.48 to 0.93.

Figure 4.- Concluded.

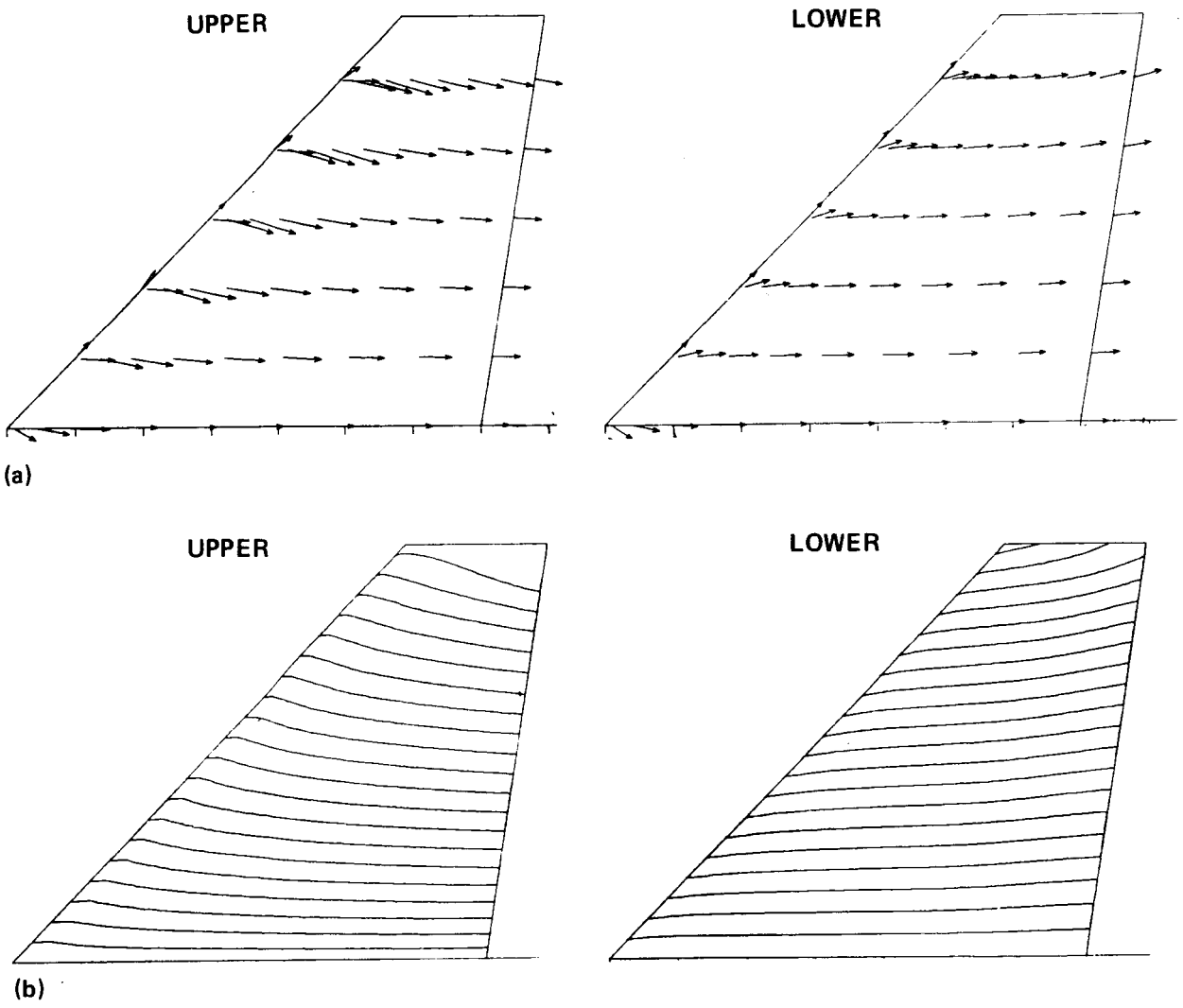
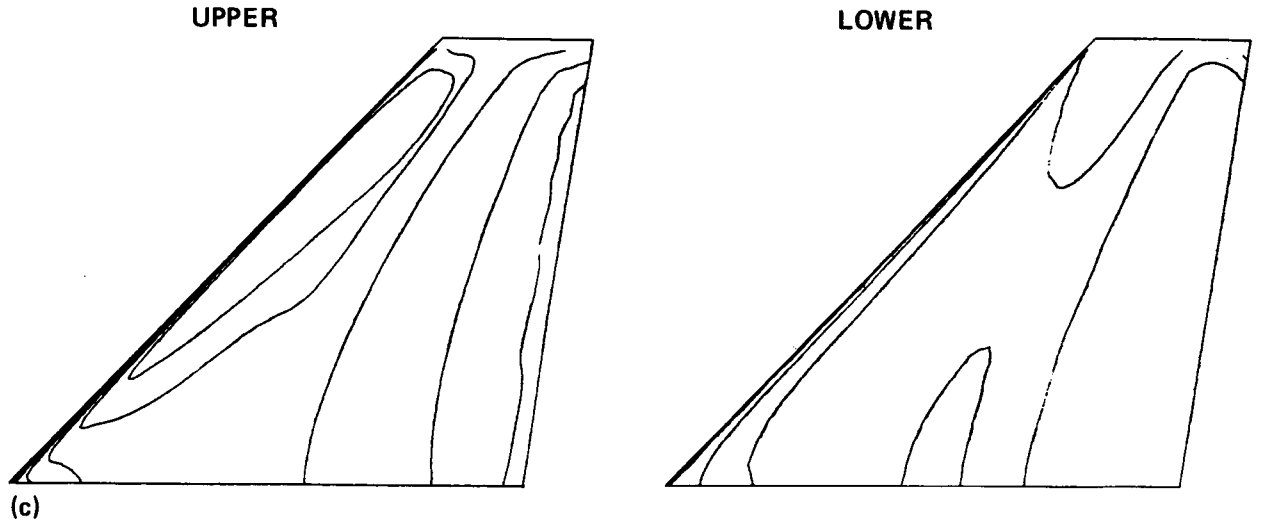


Figure 5.- Predicted inviscid surface-flow characteristics by FL022 code; M = 0.85, a = 5°. (a) Velocity vectors. (b) Streamlines.



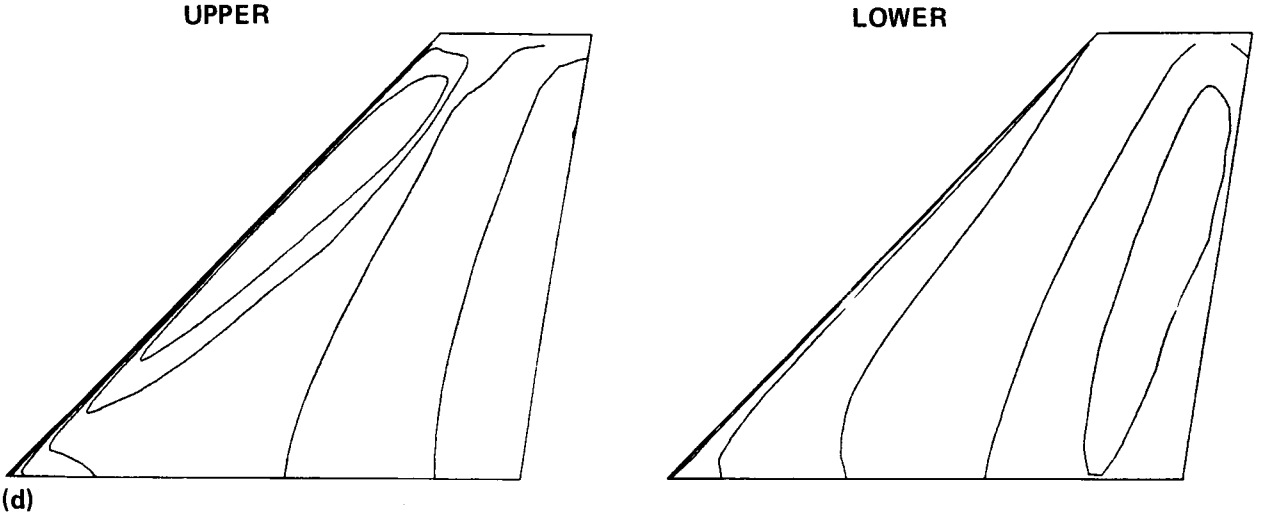


Figure 5.- Concluded. (c) Pressure contours. (d) Mach number contours.

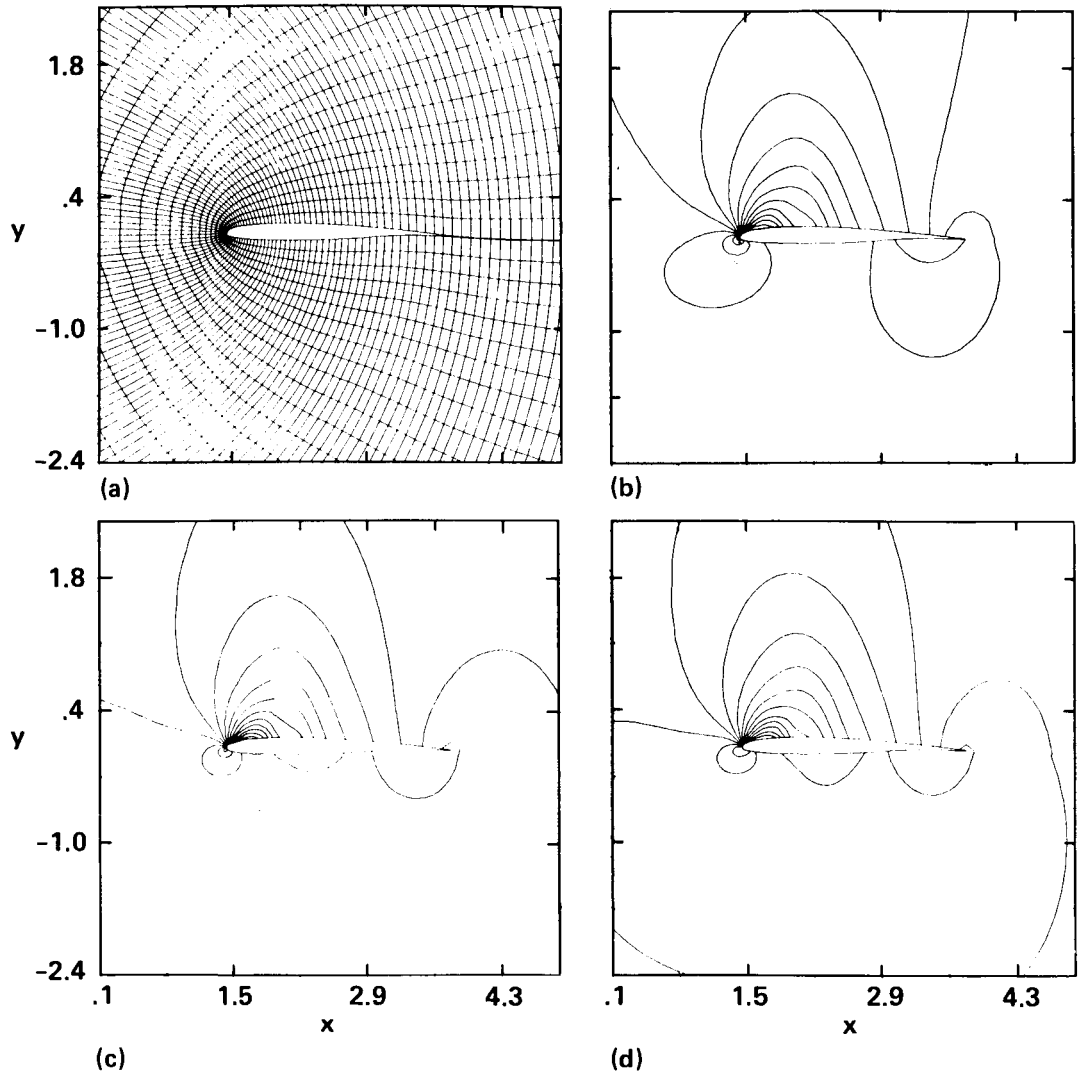


Figure 6.- Predicted inviscid flow characteristics in a vertical, chordwise plane by FL022 code; M = 0.85, a = 5°, n = 0.488. (a) Grid. (b) Pressure contours. (c) Mach number contours. (d) Density contours.

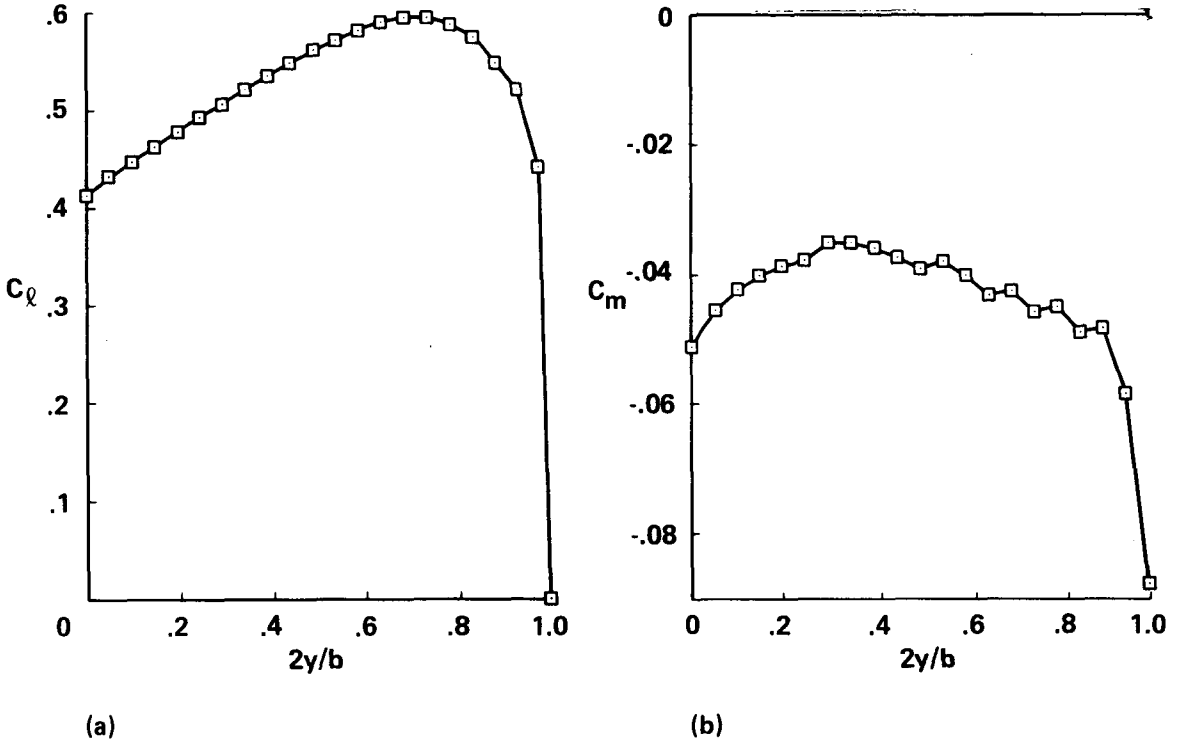
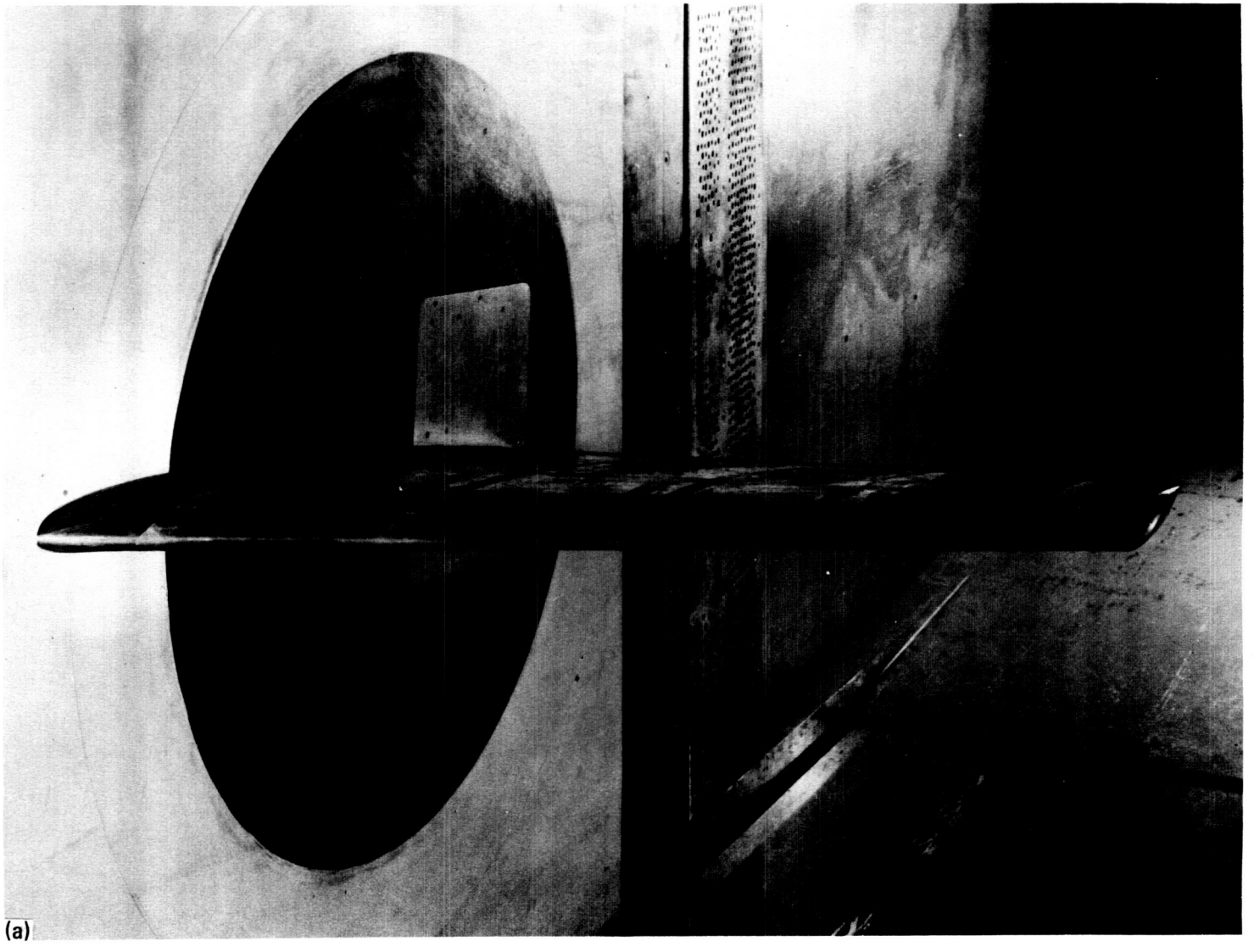
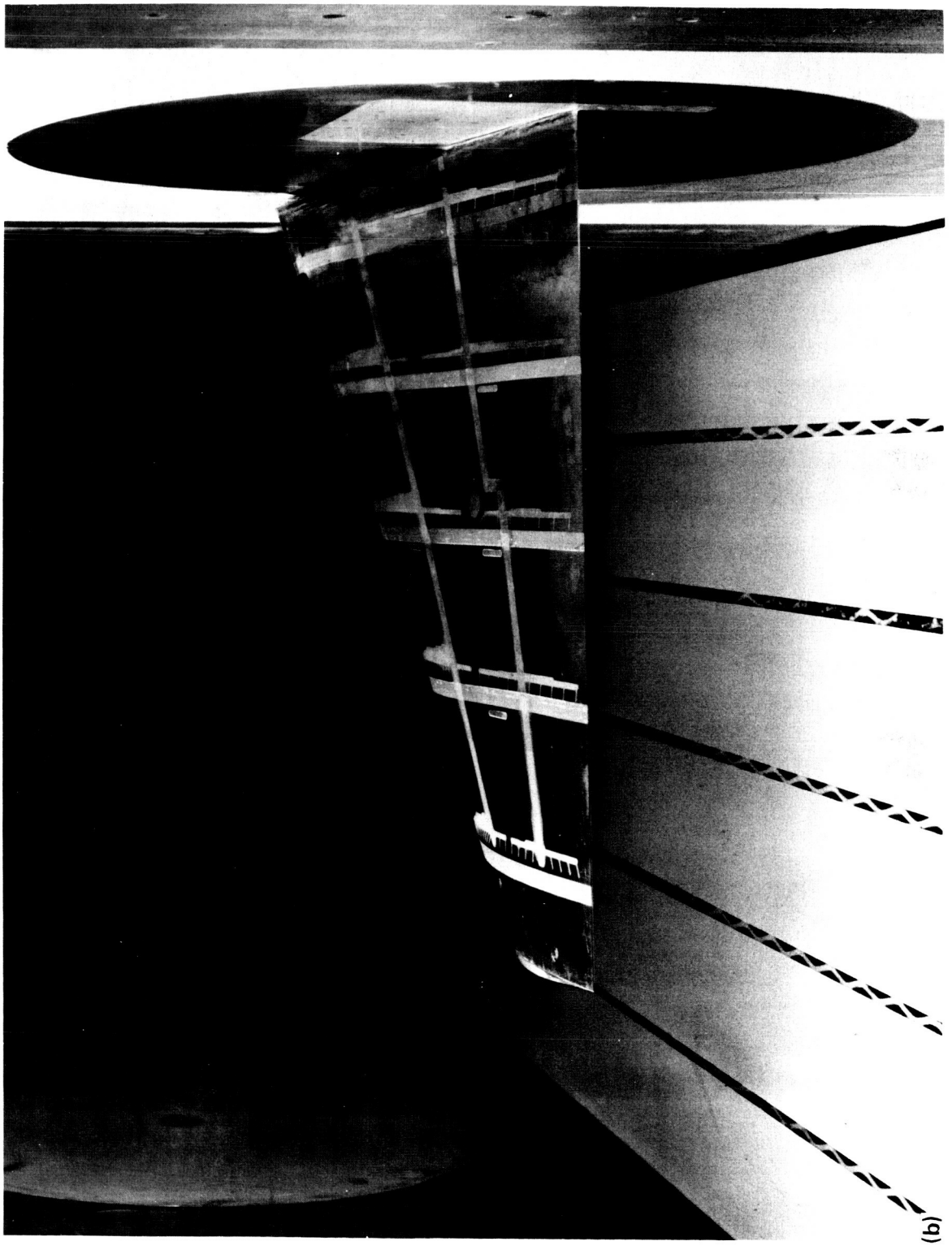


Figure 7.- Predicted spanwise loading and moments from FL022 code; M = 0.85, a = 5°. (a) Section load coefficient. (b) Section quarter-chord pitching-moment coefficient.



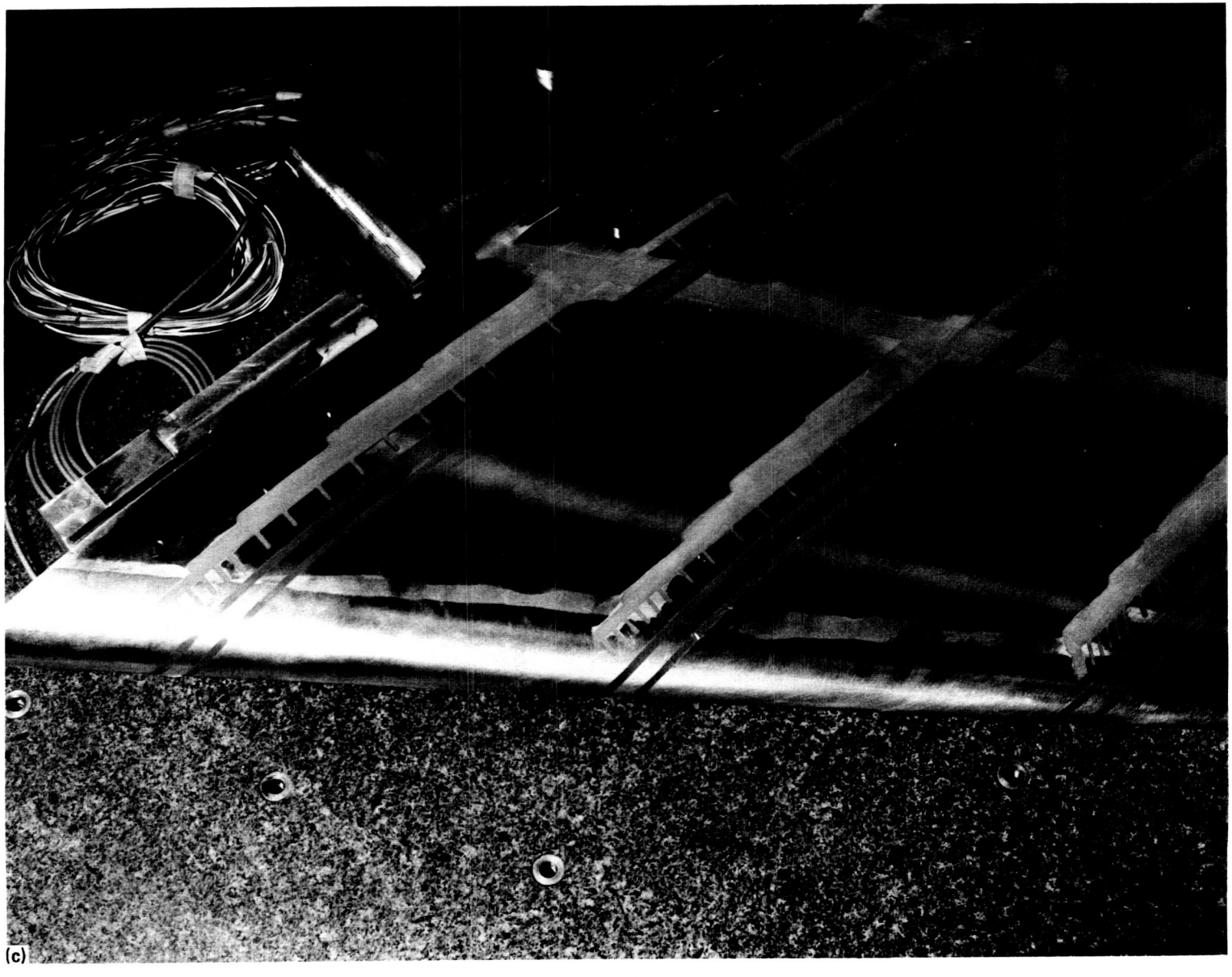
(a) Front view.

Figure 8.- Photographs of 0.9 m semispan model of Wing C mounted on wall of Ames 6- by 6-Foot Transonic Wind Tunnel.



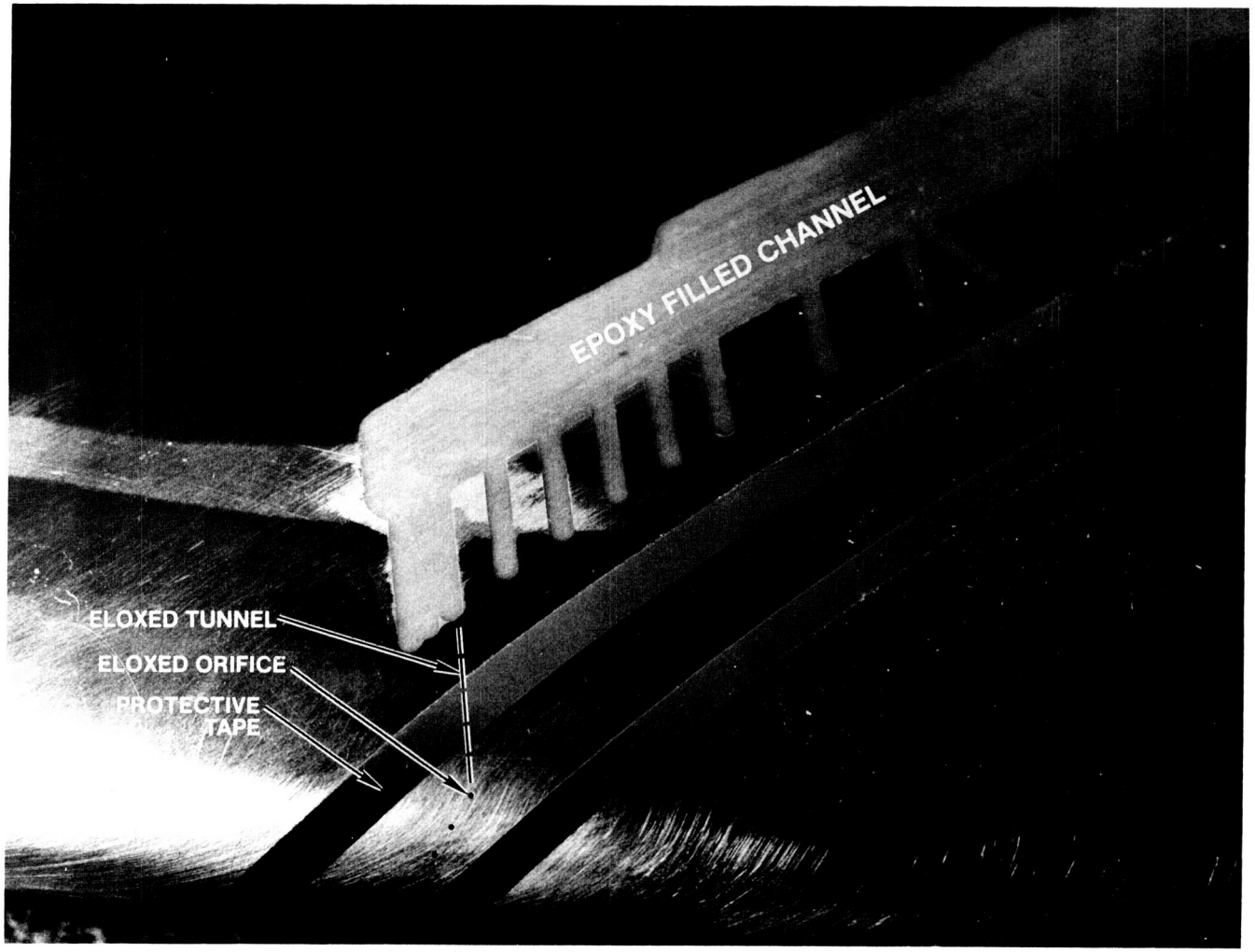
(b) Rear view.

Figure 8.- Continued.



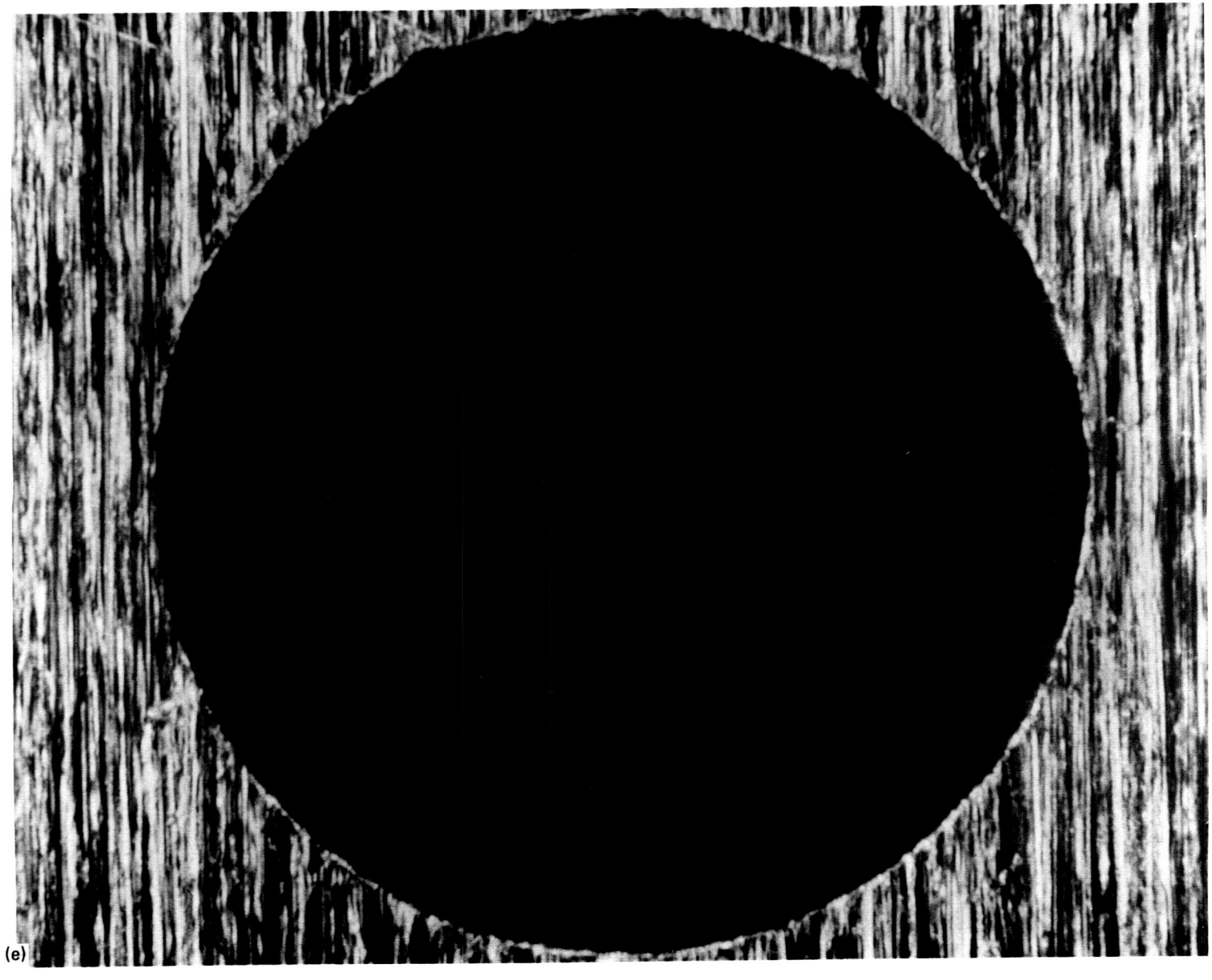
(c) Pressure tube installation.

Figure 8.- Continued.



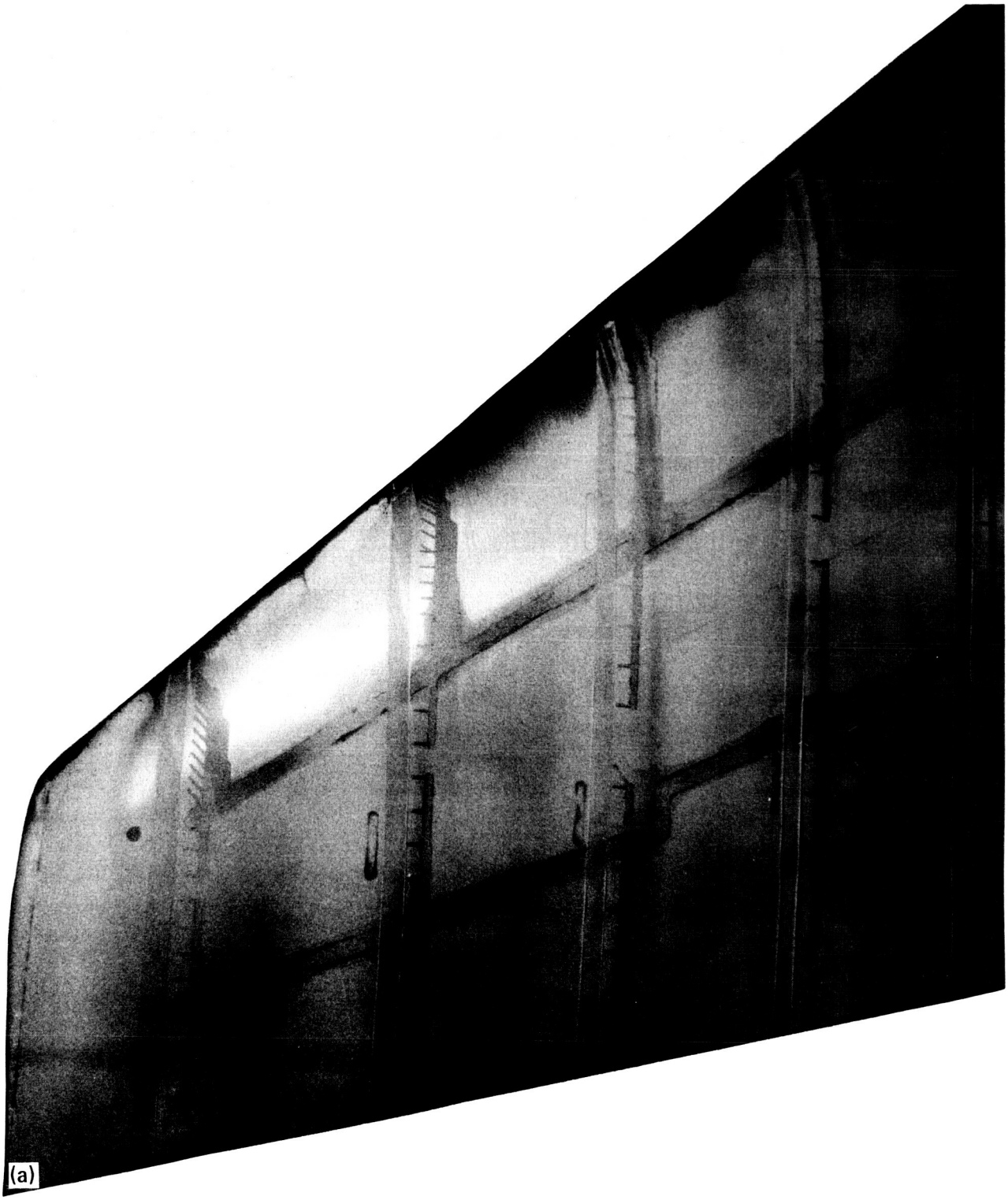
(d) Close-up view of pressure tube installation.

Figure 8.- Continued.



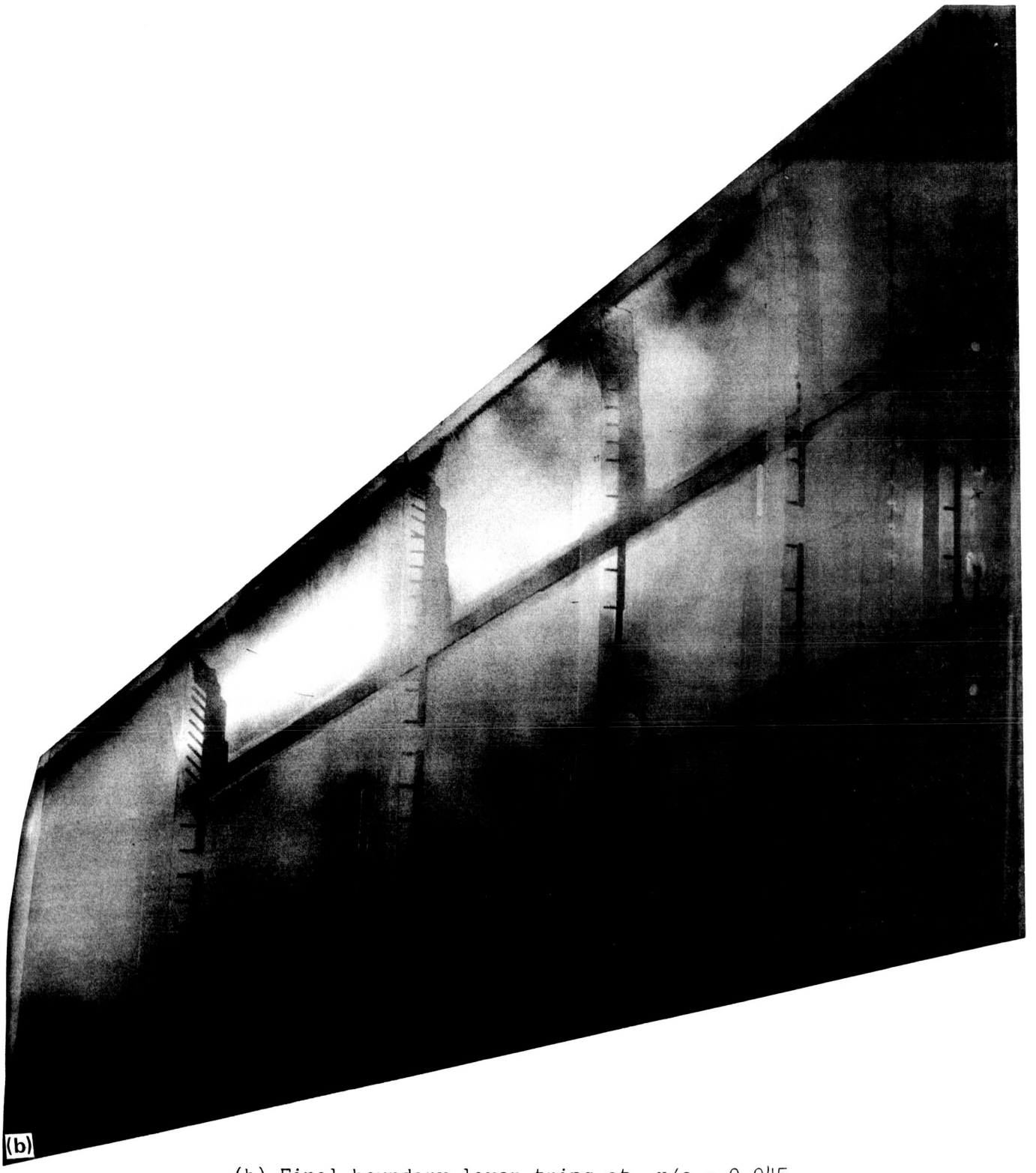
(e) Enlarged view of typical EDM pressure orifice (0.50 mm diam).

Figure 8.- Concluded.



(a) No boundary-layer trips, natural transition.

Figure 9.- Photographs of sublimation tests for the location of boundary-layer transition; M = 0.85, a = 5°, Re = 10×10^6.

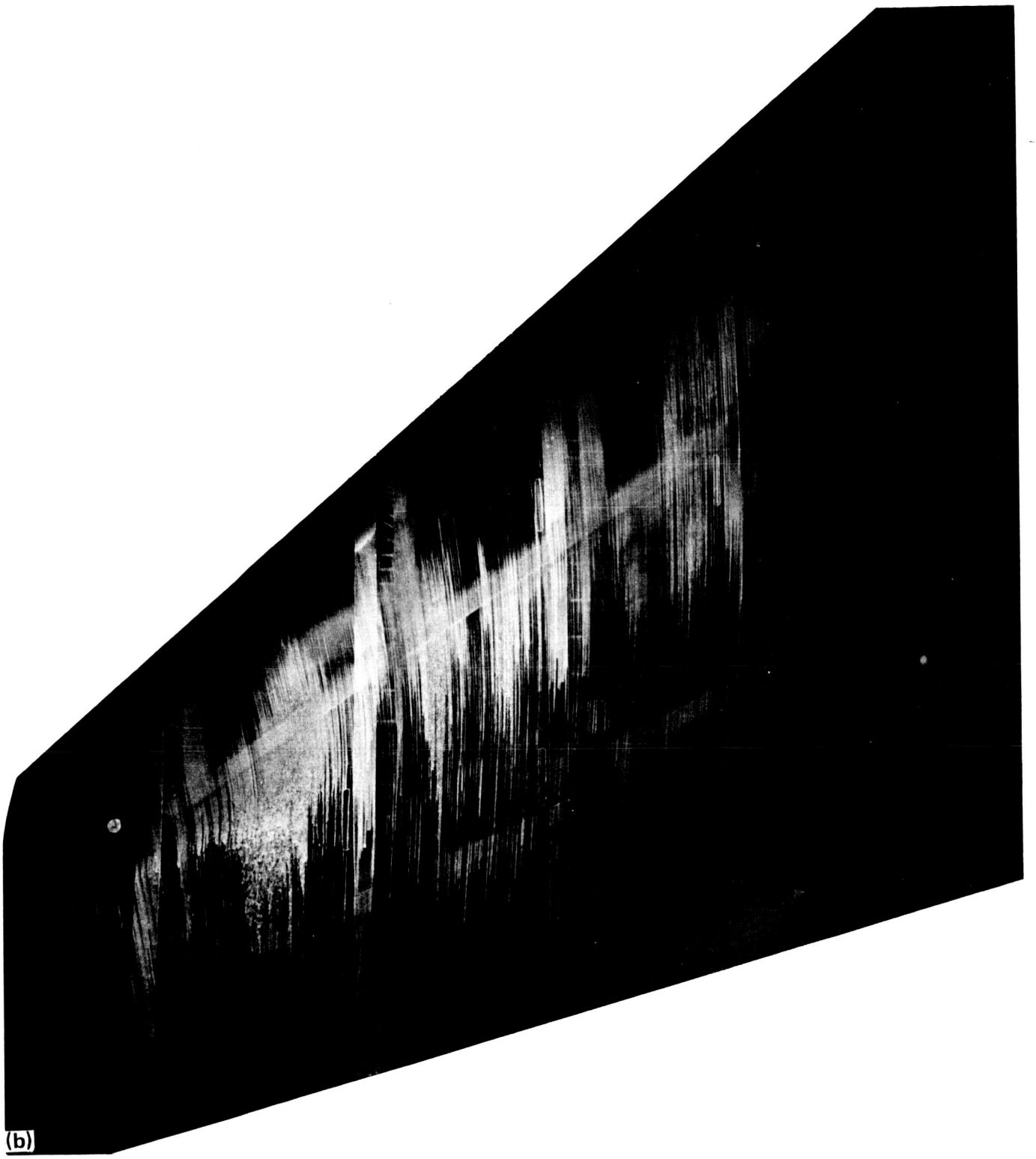


(b) Final boundary-layer trips at x/c = 0.045.

Figure 9.- Concluded.

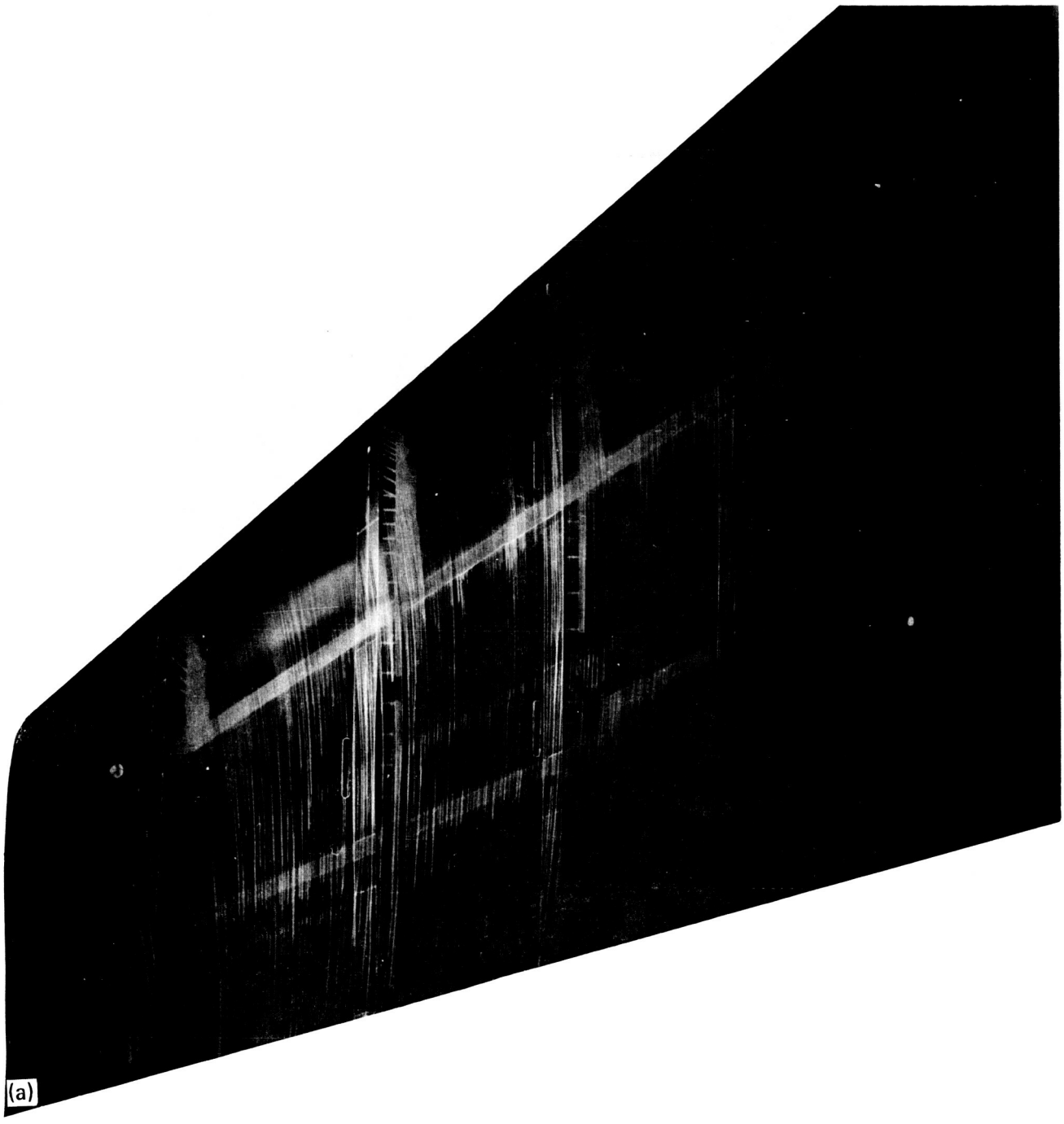
(a) M = 0,85, design test condition.

Figure 10.- Oil-flow photographs; a = 5°, Re = 10×10^6.



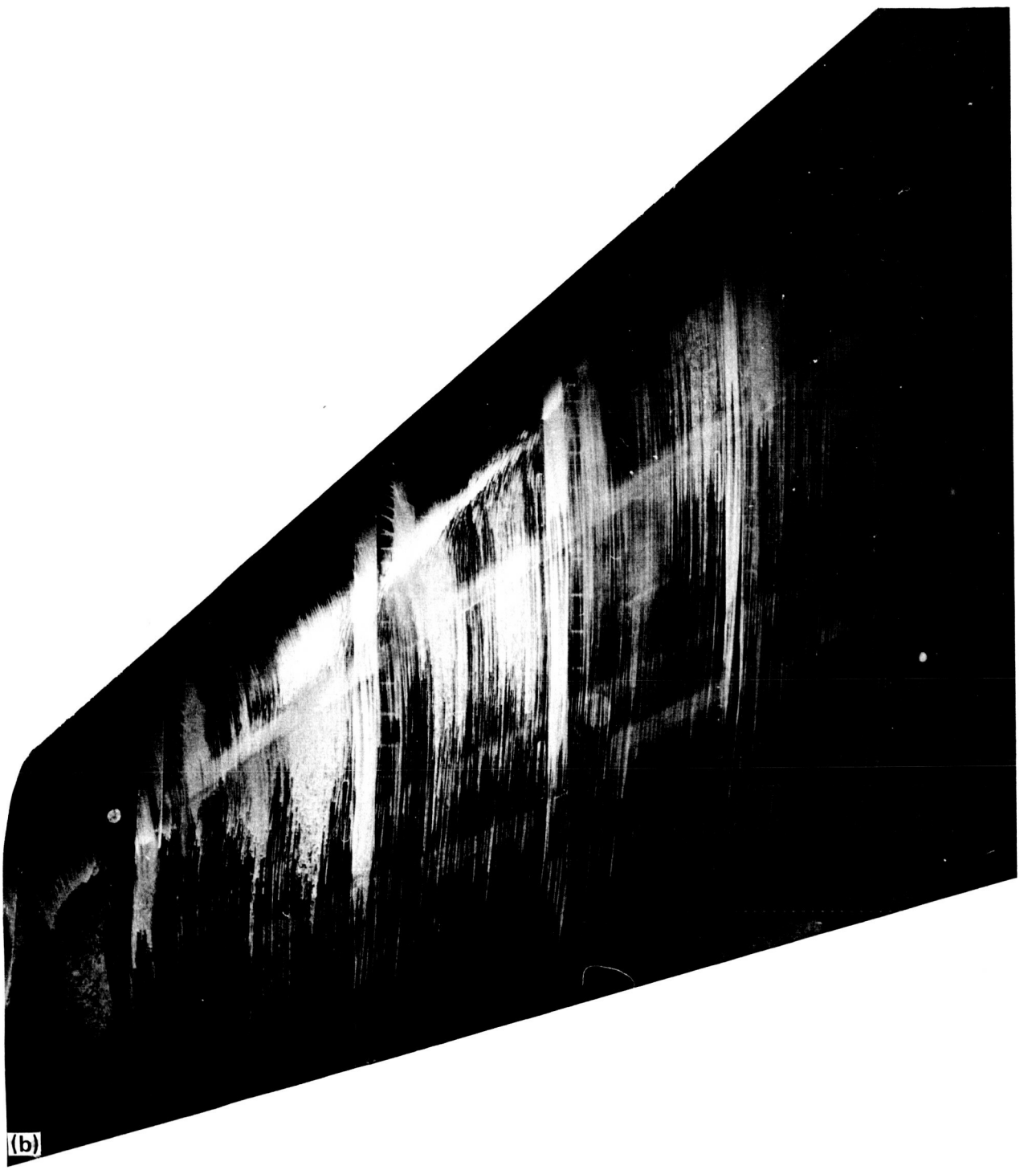
(b) M = 0.82.

Figure 10.- Concluded.



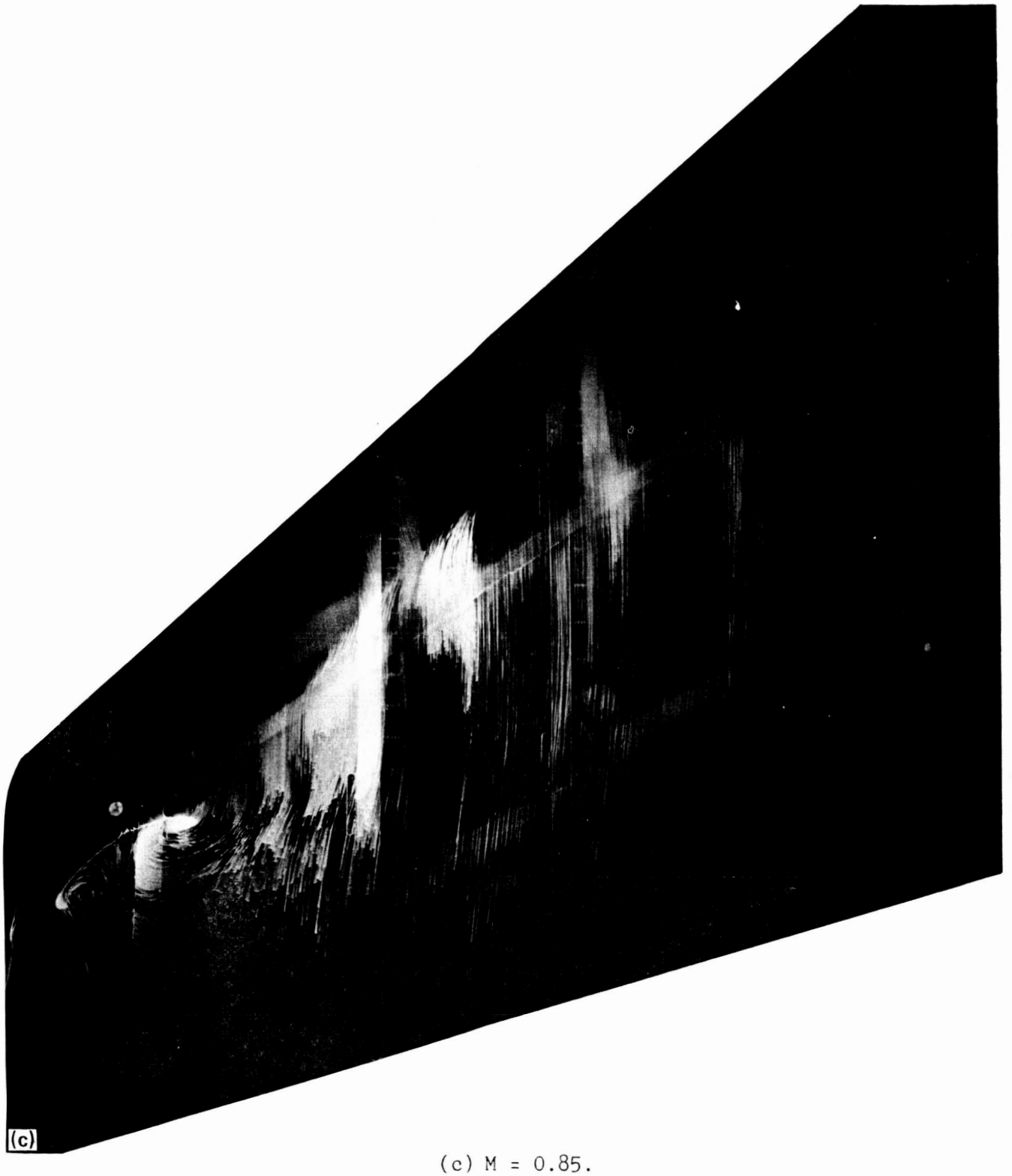
(a) M ▫ 0.70.

Figure 11.- Oil-flow photographs; a = 5°, Re = 6.8×10^6.



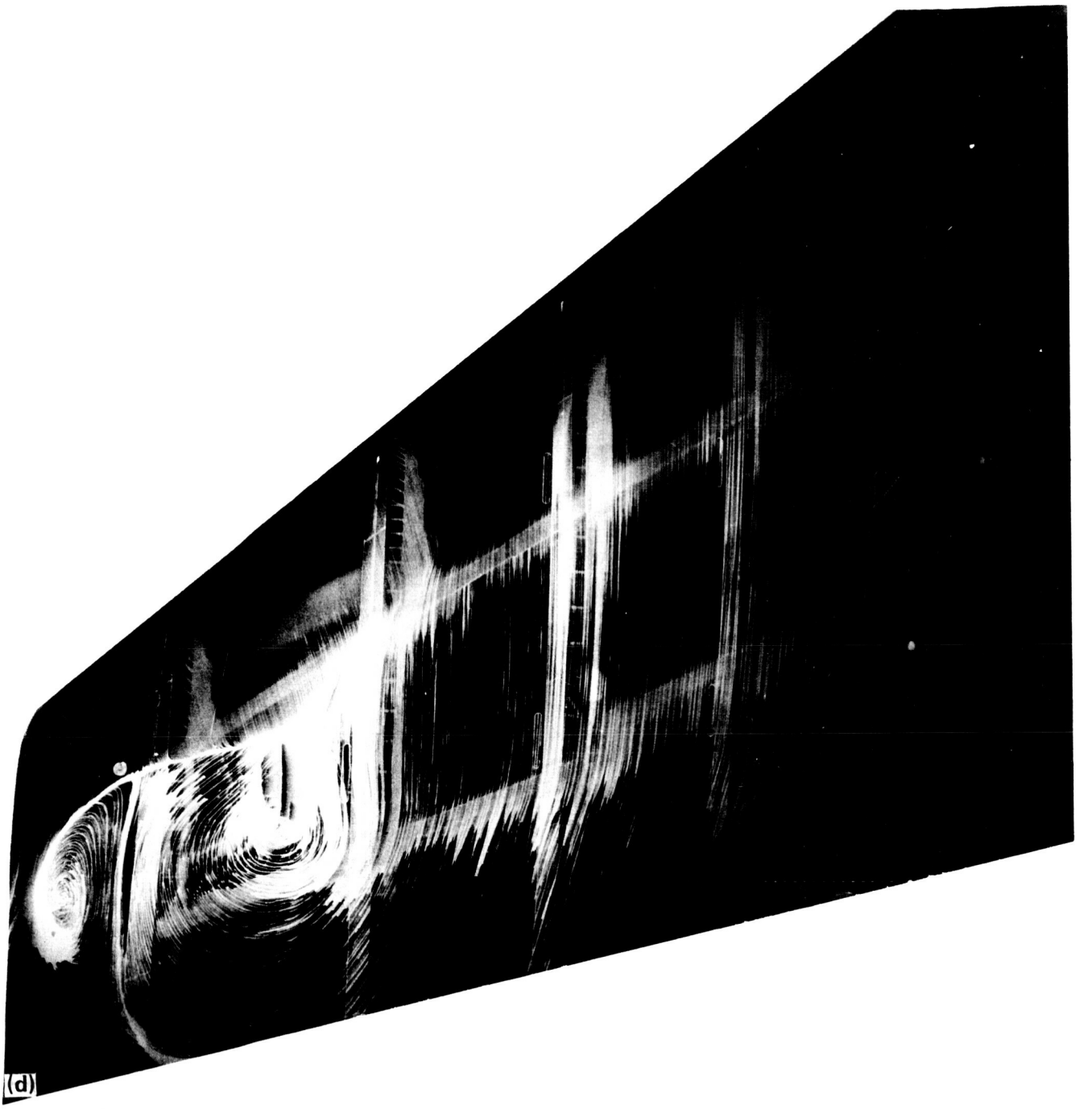
(b) M = 0.82.

Figure 11.- Continued.



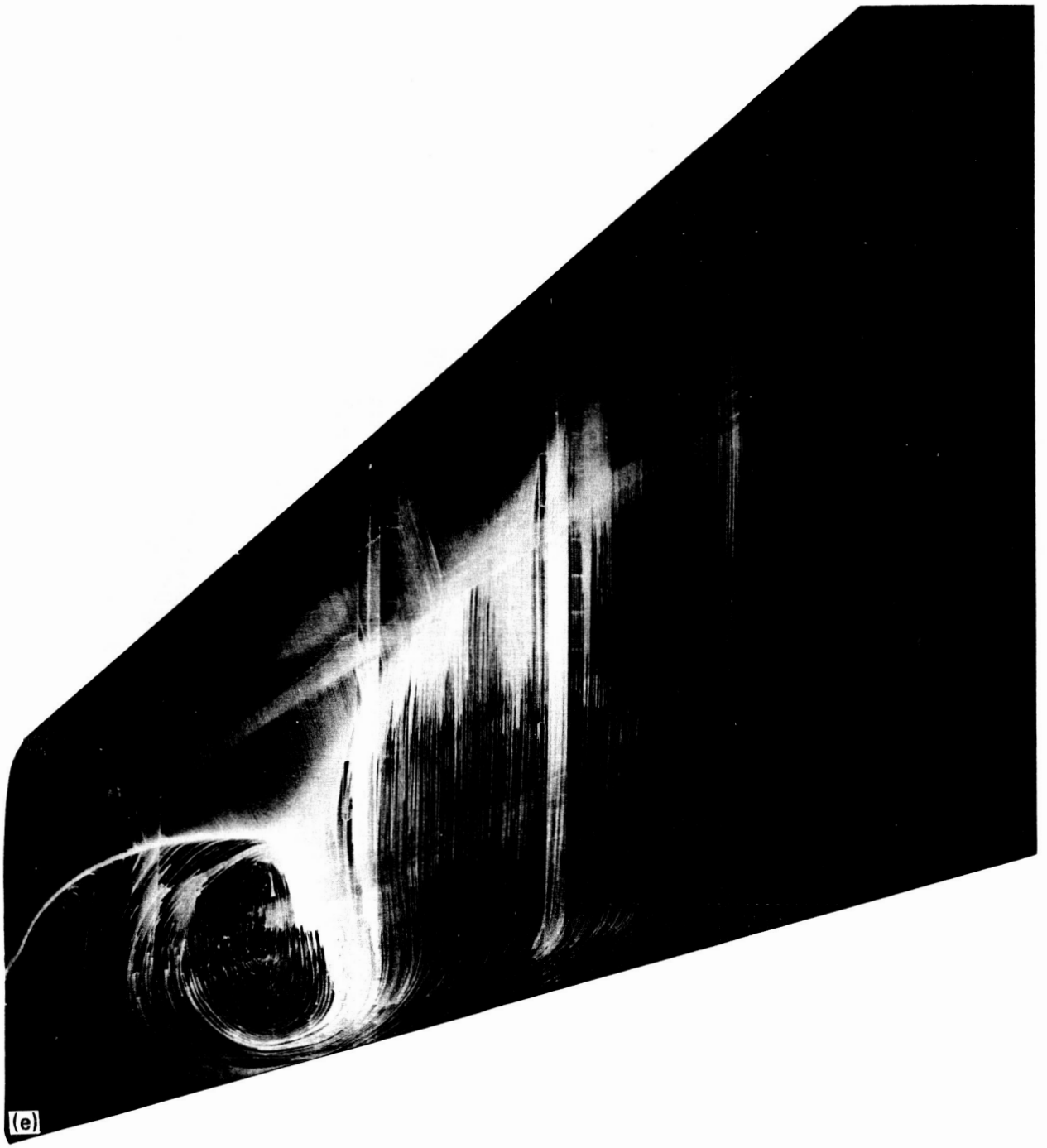
(c) M = 0.85.

Figure 11.- Continued.



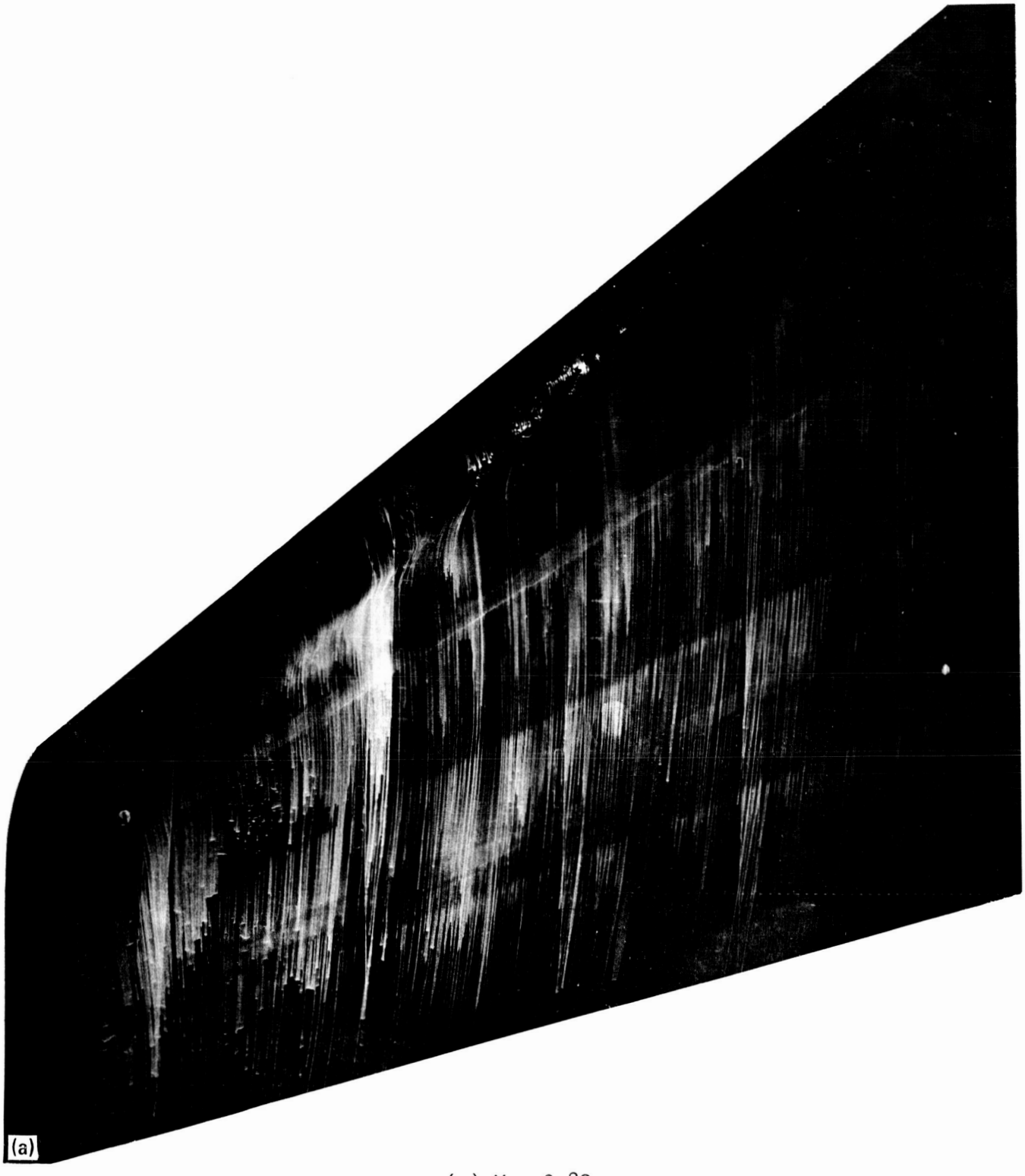
(d) M = 0.90.

Figure 11.- Continued.



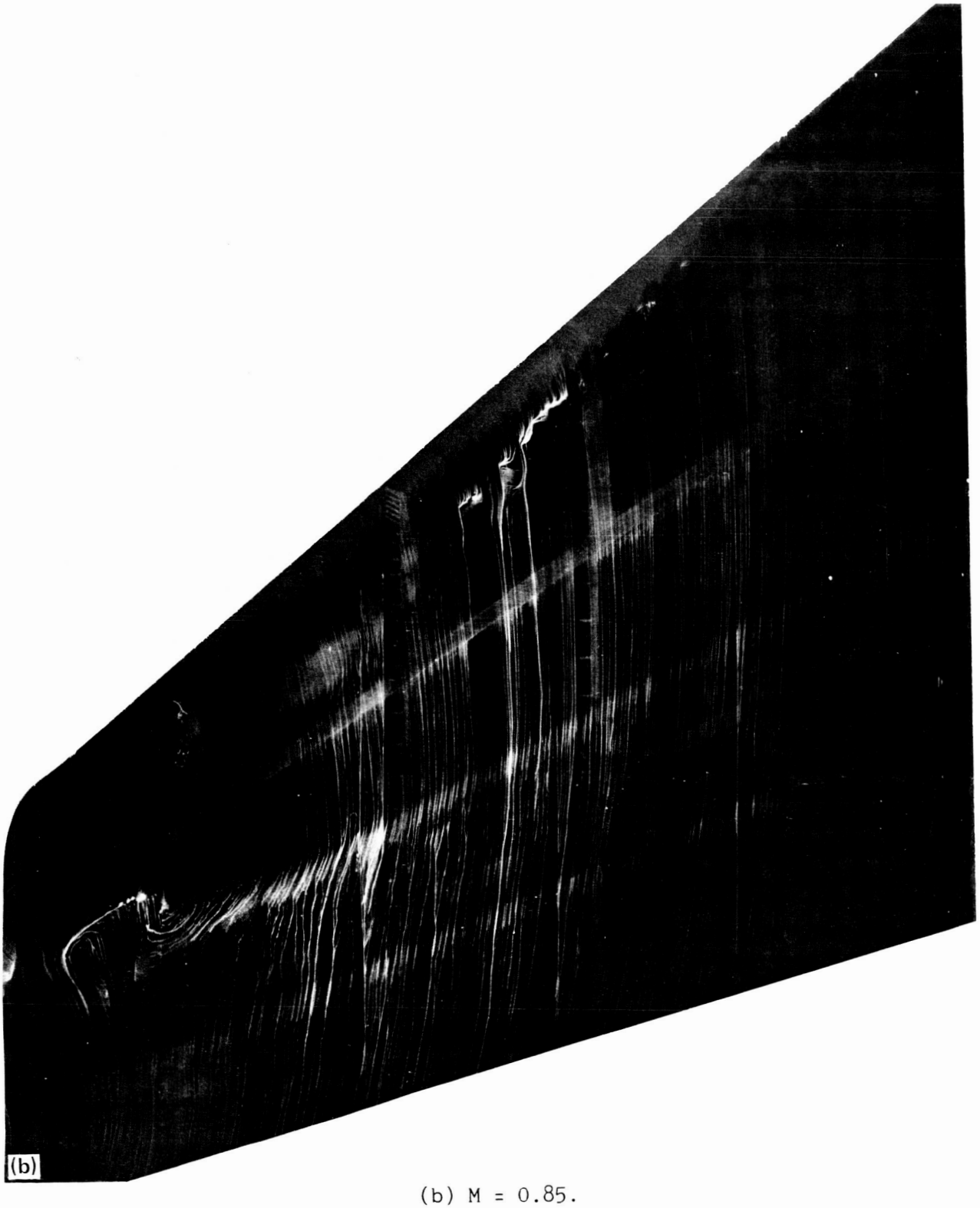
(e) M = 0.95.

Figure 11.- Concluded.



(a) M = 0.82.

Figure 12.- Oil-flow photographs: a = 5°, Re = 3.4×10^6.



(b) M = 0.85.

Figure 12.- Concluded.

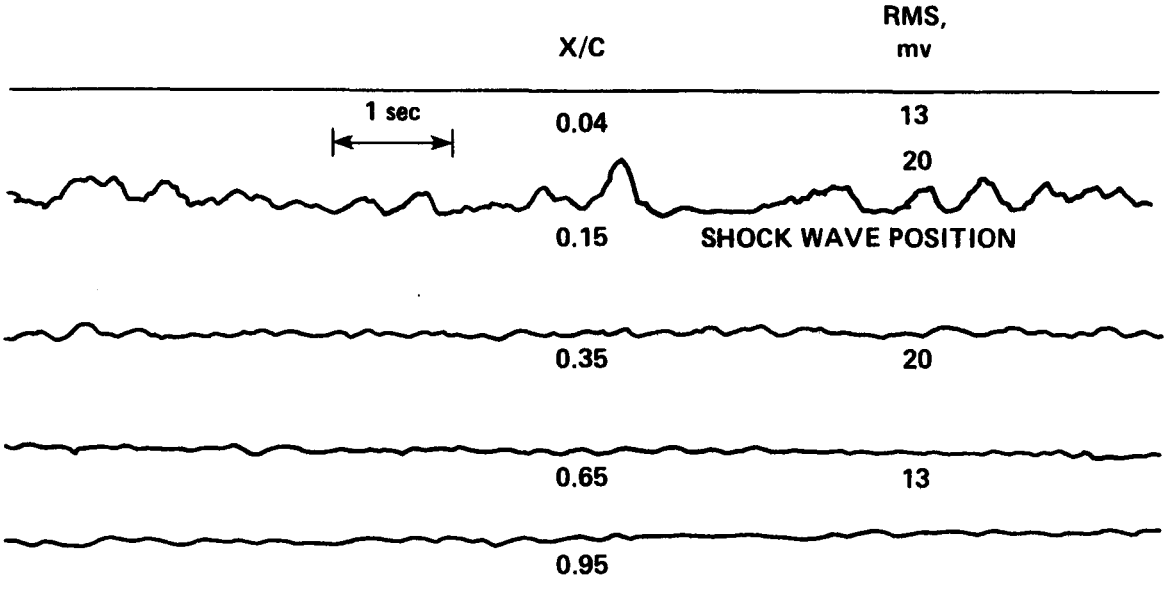


Figure 13.- Oscillograph traces of shock-wave-induced unsteady pressures at several chordwise stations; M = 0.82, a = 5°, Re = 6.8×10^6, n = 0.50.

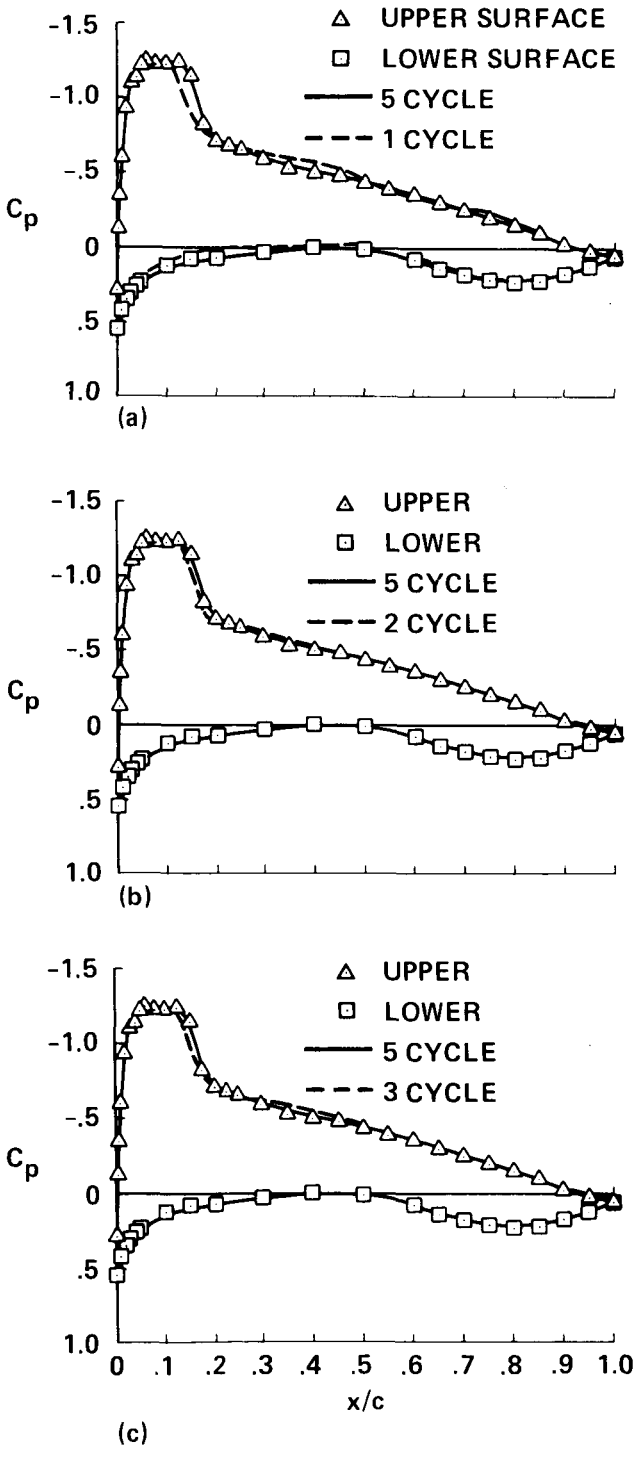
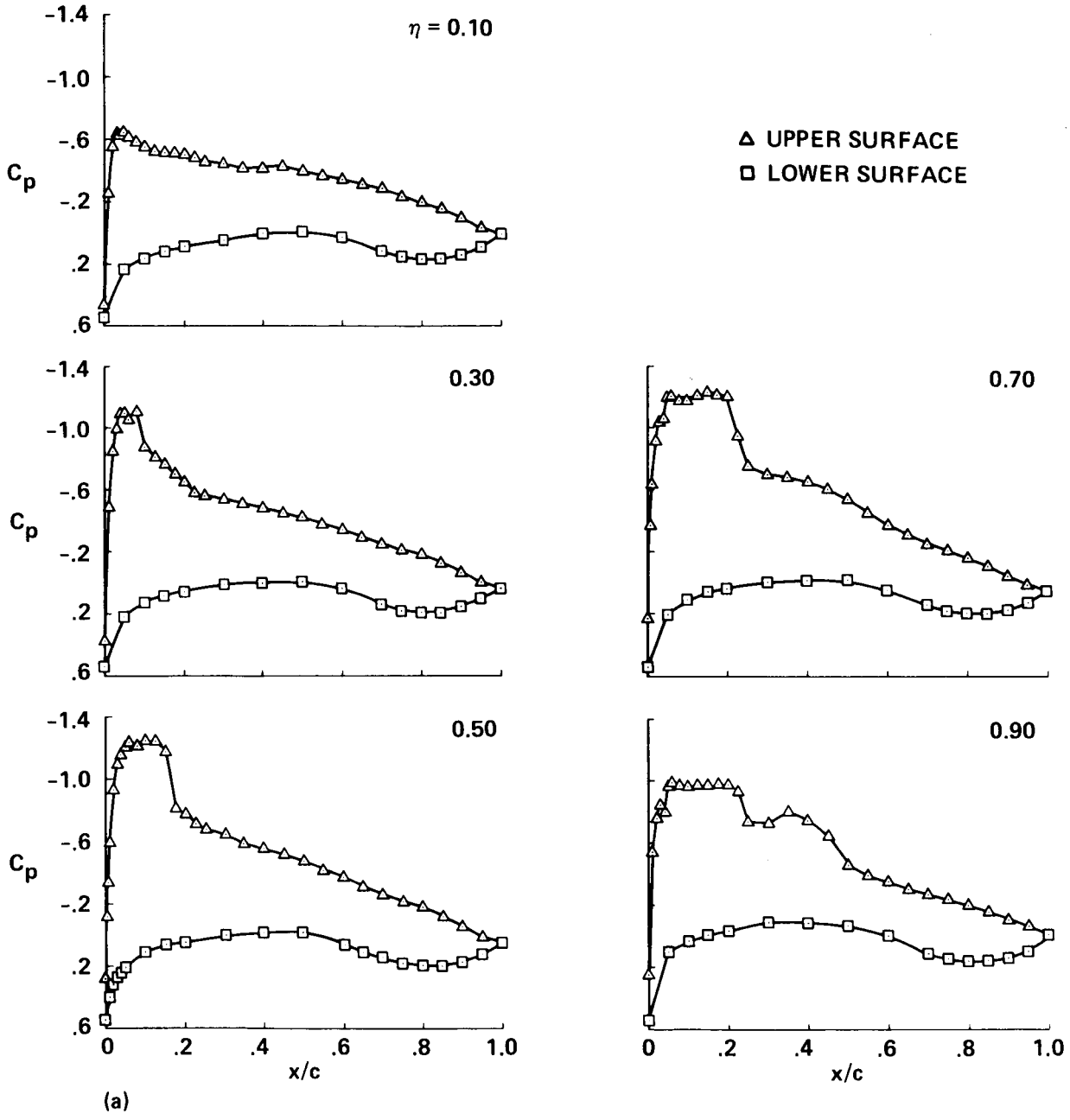
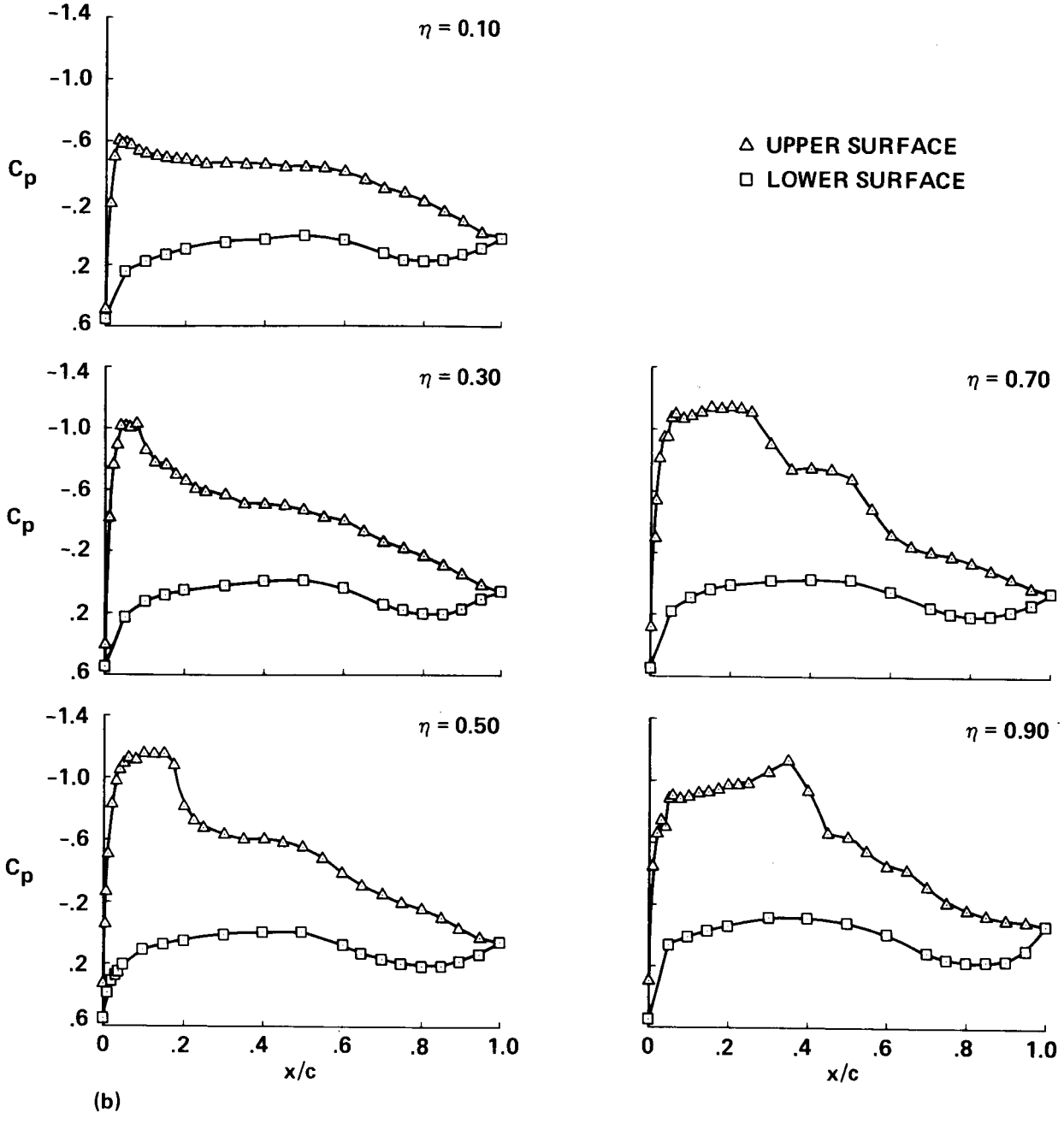


Figure 14.- Effect of multiple-cycle pressure-data averaging on experimental chordwise pressure distributrion; M = 0.82, a = 5°, Re = 6.8×10^6, n = 0.50. (a) One cycle. (b) Two cycles. (c) Three cycles.



(a) M = 0.82.

Figure 15.- Experimental chordwise-pressure distributions at design Reynolds number of 10×10^6; a = 5°.



(b) M = 0.85.

Figure 15.- Concluded.

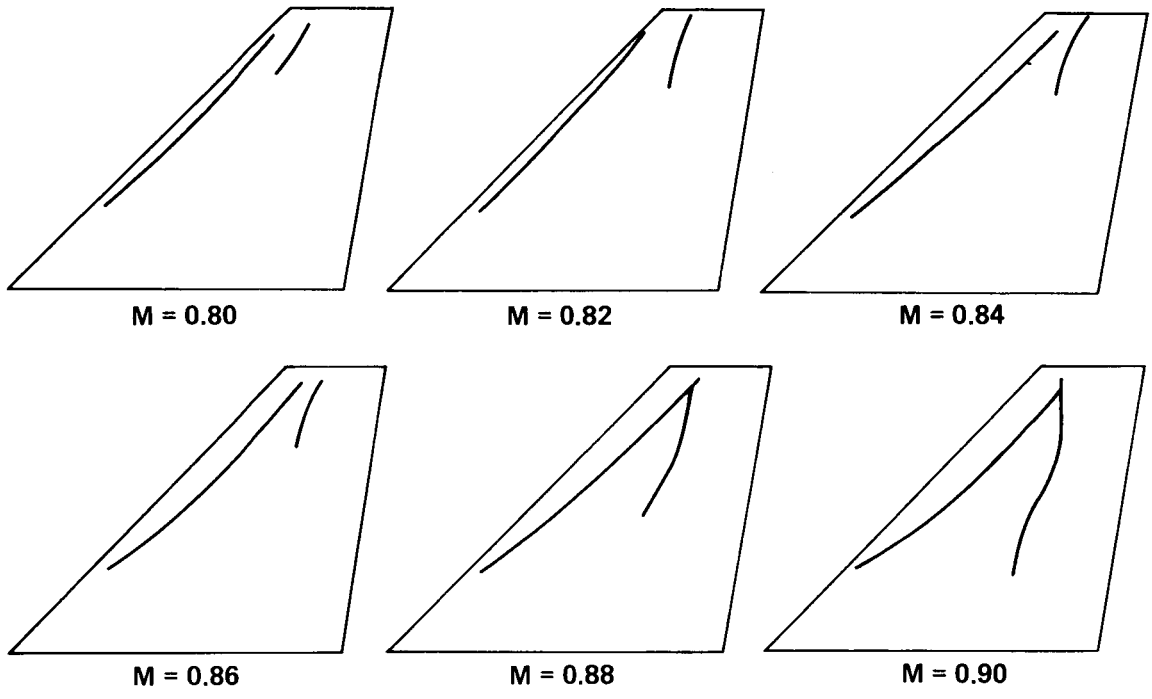
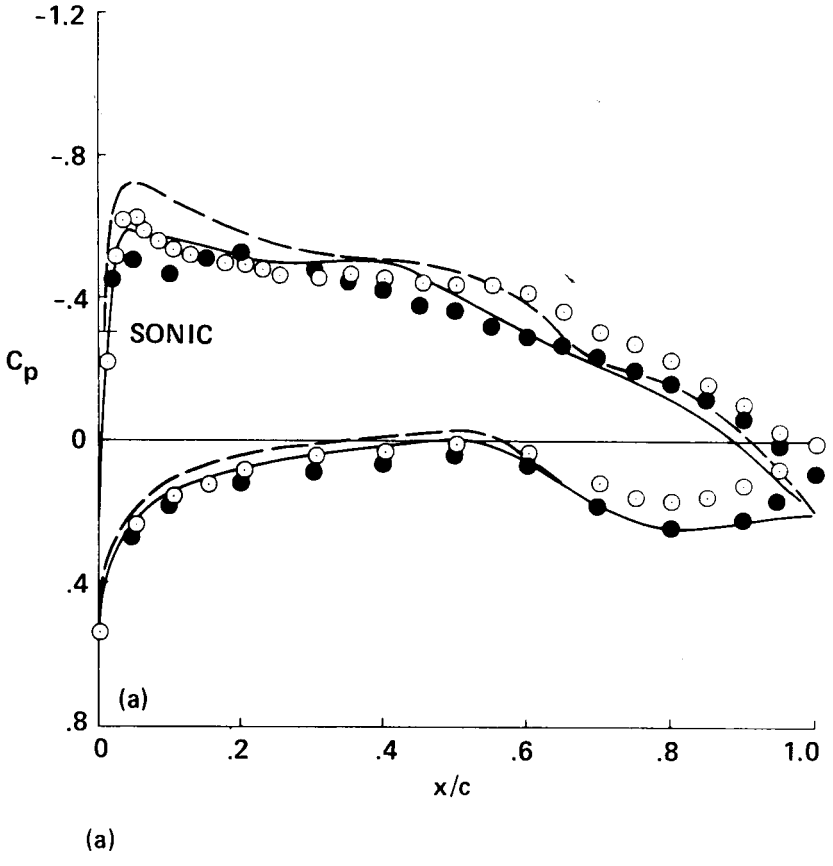


Figure 16.- Shock-wave patterns at various Mach numbers at an angle of attack of 5° (taken from ref. 3).

PEAK C_p	PEAK M	M_N			α, deg	C_N
			○	LARGE SCALE	5.0	0.54
-0.62	1.18	0.83	●	SMALL SCALE (REF. 3)	5.9	0.54
-0.50	1.11	0.78	——	FLO-22 CODE	5.0	0.52
-0.60	1.16	0.82	— —	TWING (REF. 13)	5.0	0.52

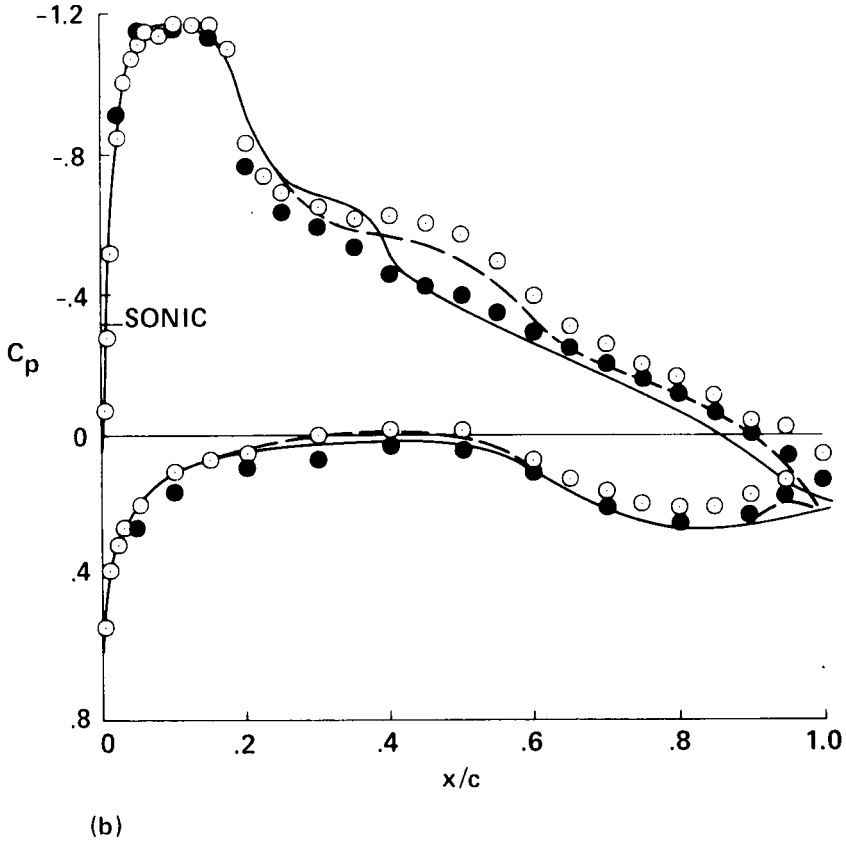


(a)

(a) n = 0.10.

Figure 17.- Comparison of experimental and predicted chordwise-pressure distributions for design conditions: M = 0.85, a = 5°, Re = $10 \times 10_6$.

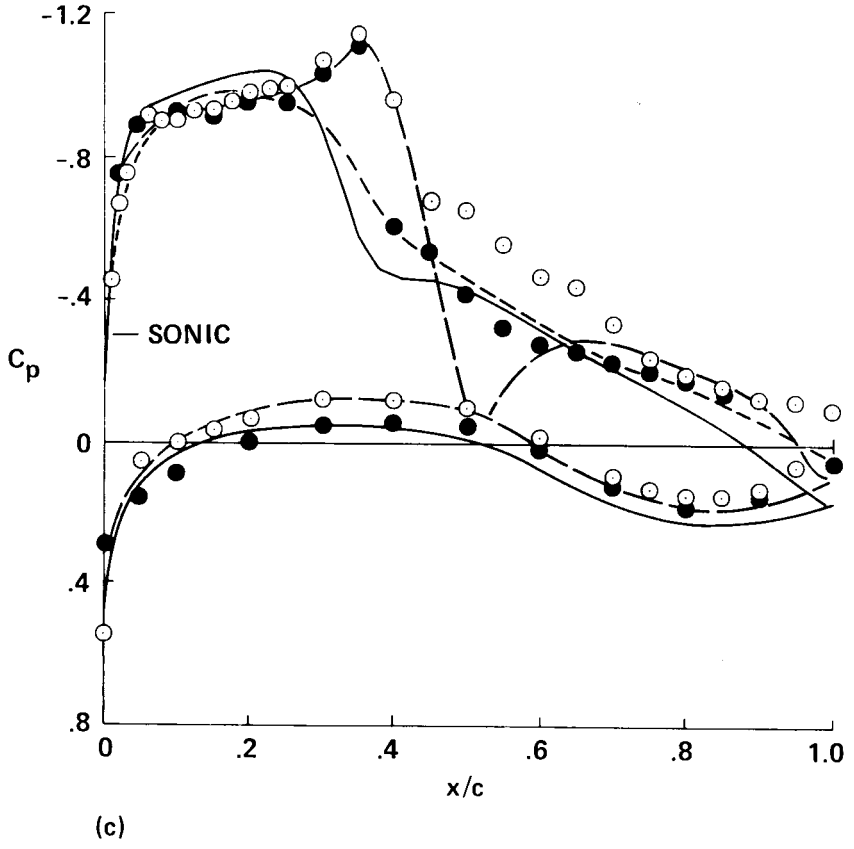
PEAK C_p	PEAK M	M_N		α, deg	C_N
			○ LARGE SCALE	5.0	0.54
-1.17	1.56	1.10	● SMALL SCALE (REF. 3-5)	5.9	0.54
-1.17	1.56	1.10	—— FLO-22 CODE	5.0	0.52
-1.17	1.56	1.10	— — TWING CODE (REF. 13)	5.0	0.52



(b) n = 0.50.

Figure 17.- Continued.

PEAK C_p	PEAK M	M_N			α, deg	C_N	η
			○	LARGE SCALE	5.0	0.54	0.90
-1.15	1.54	1.09	●	SMALL SCALE (REF. 3)	5.9	0.54	0.90
-1.13	1.52	1.07	——	FLO-22 CODE	5.0	0.52	0.88
-1.04	1.45	1.06	— —	TWING (REF. 13)	5.0	0.52	0.91
-1.00	1.41	1.00	- - - -	SPECIFIED DESIGN P.D.	5.0	0.52	0.91



(c) n = 0.90.

Figure 17.- Concluded.

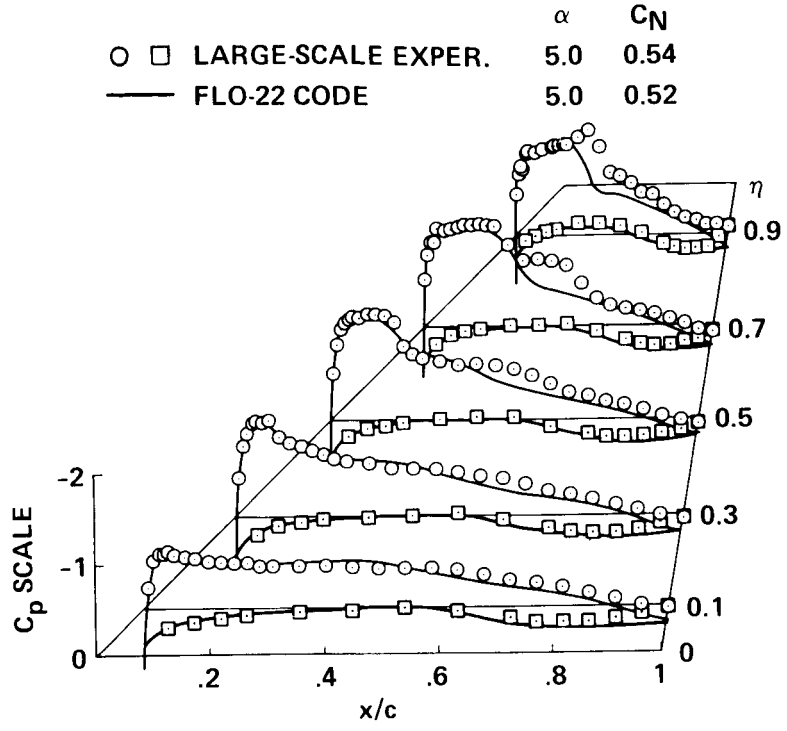


Figure 18.- Carpet-plot comparison of experimental and predicted chordwise-pressure distributions for design conditions: M = 0.85, a = 5°, Re = 10×10^6.

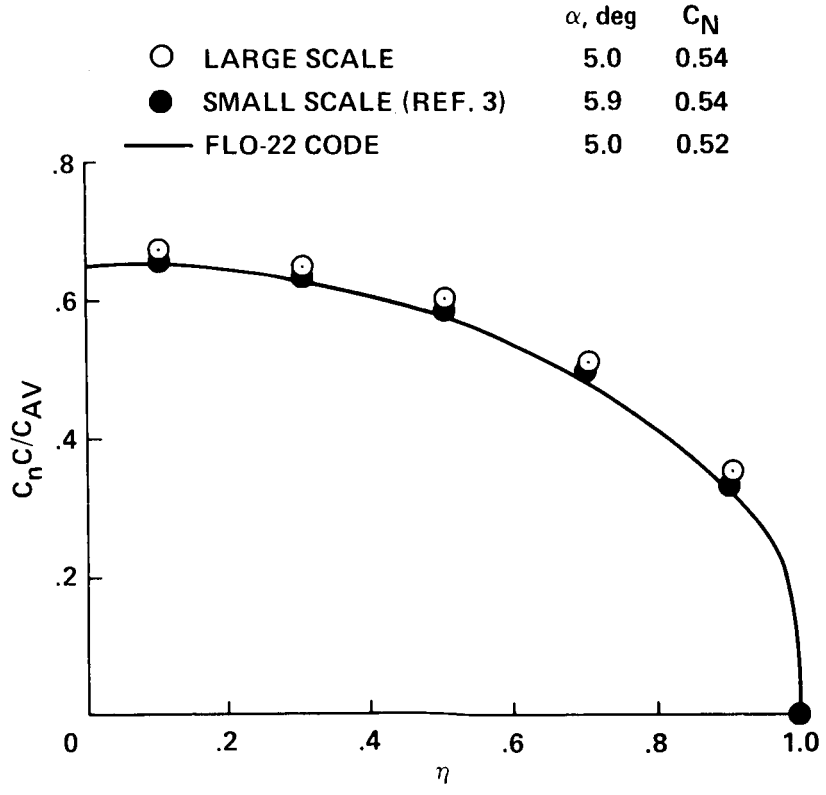


Figure 19.- Comparison of experimental and predicted spanwise load distributions for design conditions; M = 0.85, a = 5°, Re = 10×10^6.

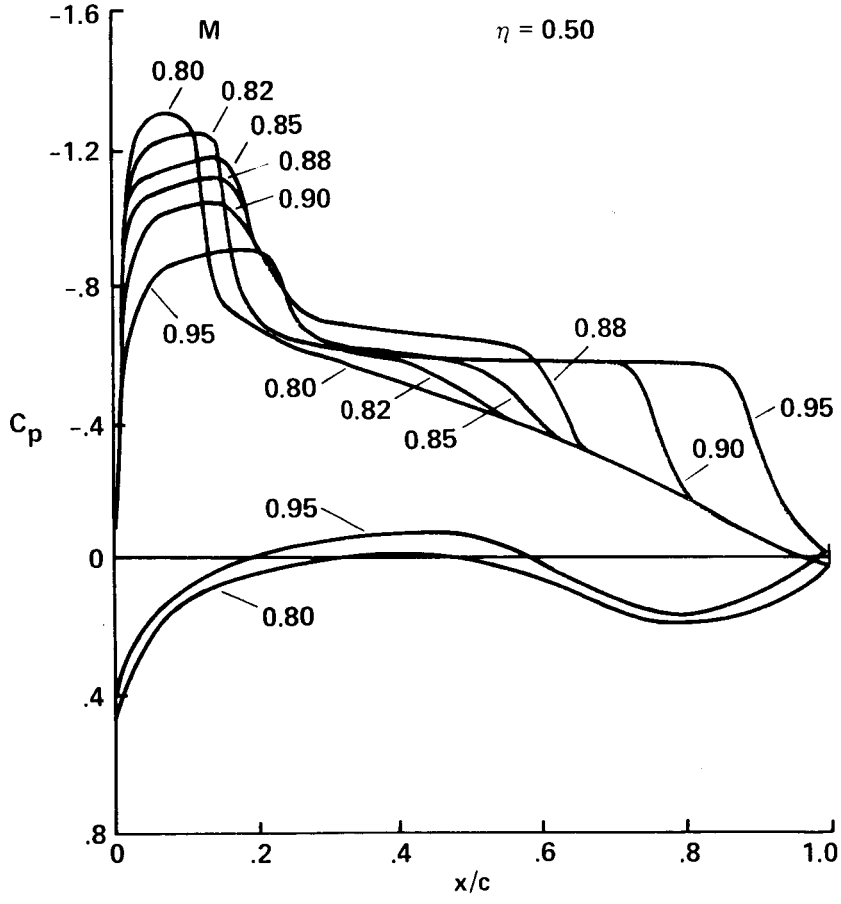


Figure 20.- Effect of Mach number on predicted chordwise pressure distributions; a = 5°, Re = 10×10^6, n = 0.50.

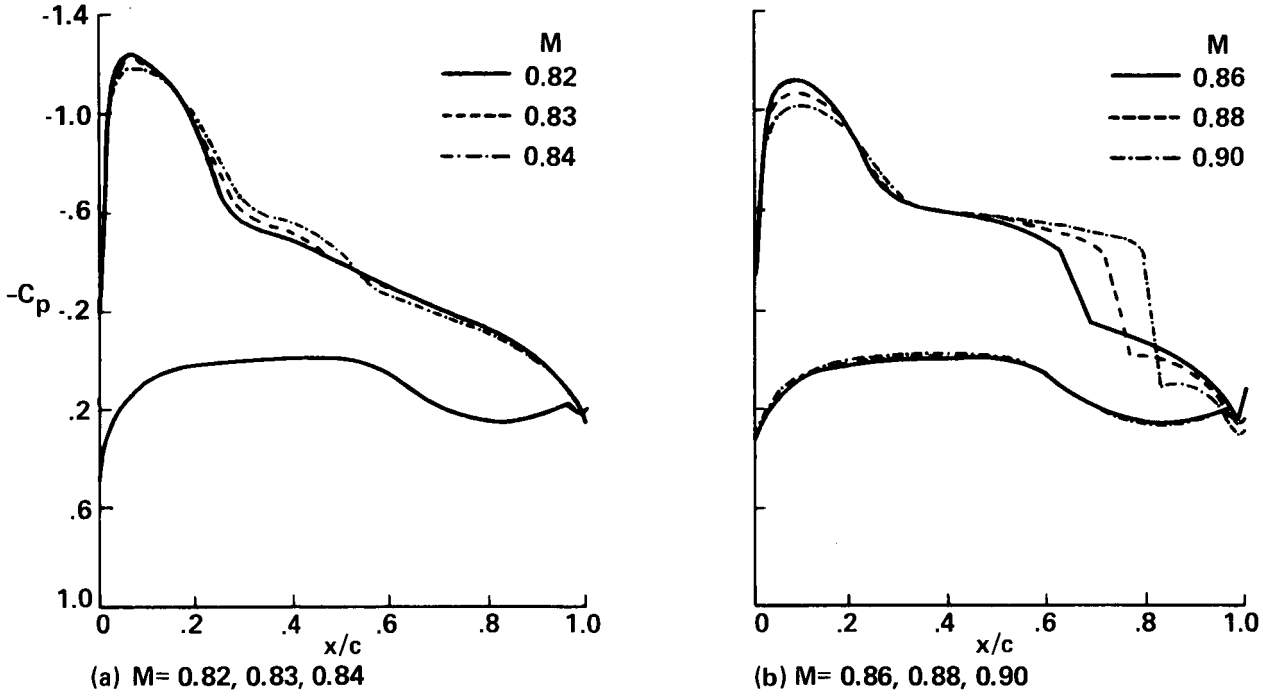


Figure 21.- Effect of Mach number on predicted chordwise pressure distributions by TWING code (refs. 10 to 12); a = 5°, n = 0.51.

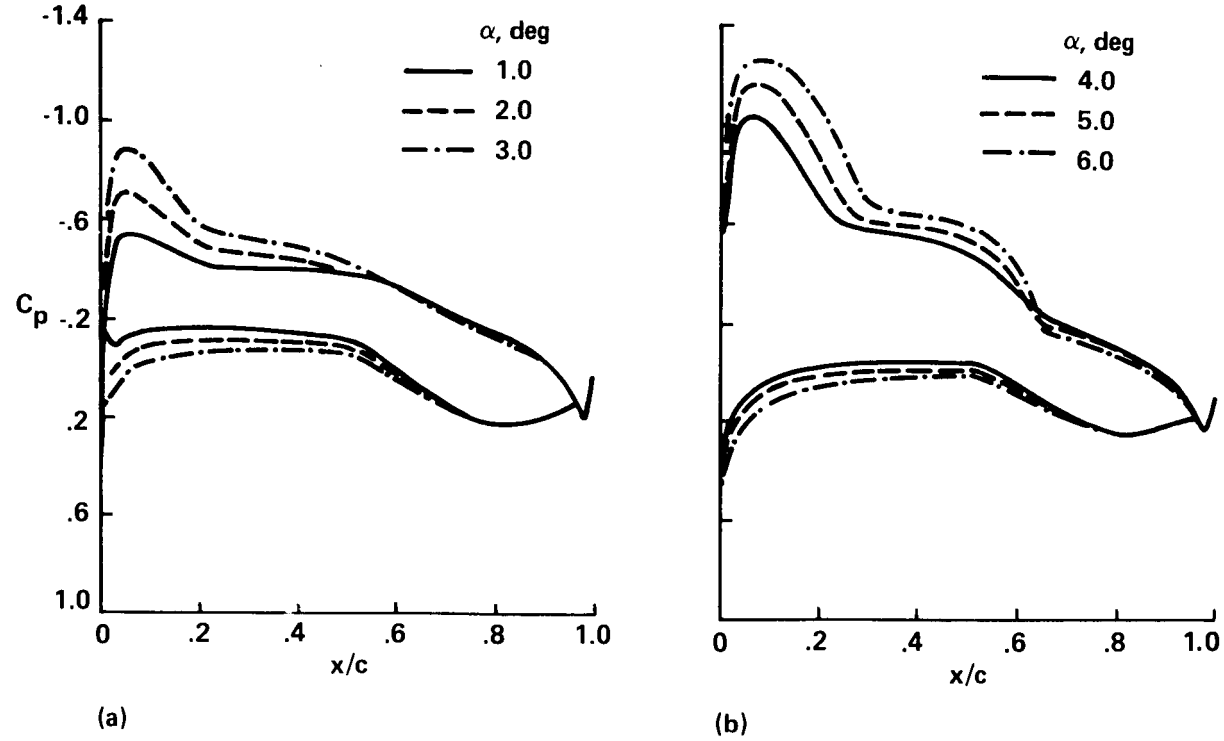


Figure 22.- Effect of angle of attack on predicted chordwise pressure distributions by TWING code (refs. 10 to 12); M = 0.85, n = 0.51. (a) a = 1°, 2°, and 3°. (b) a = 4°, 5°, and 6°.

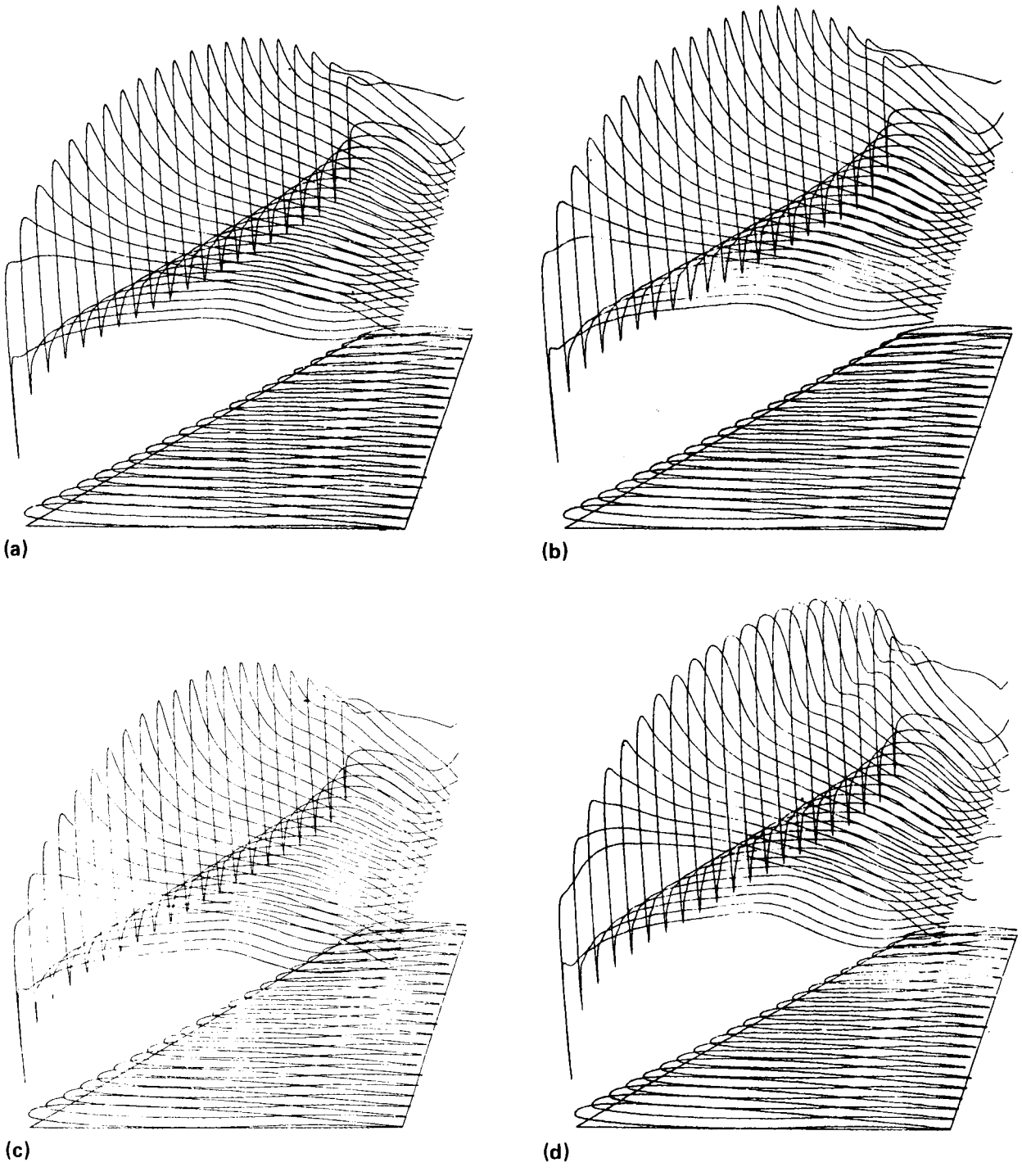


Figure 23.- Carpet plots of predicted chordwise pressure distributions by FL022 code for off-design conditions, a = 5° (except (e)). (a) M = 0.25. (b) M = 0.50. (c) M = 0.70. (d) M = 0.82.

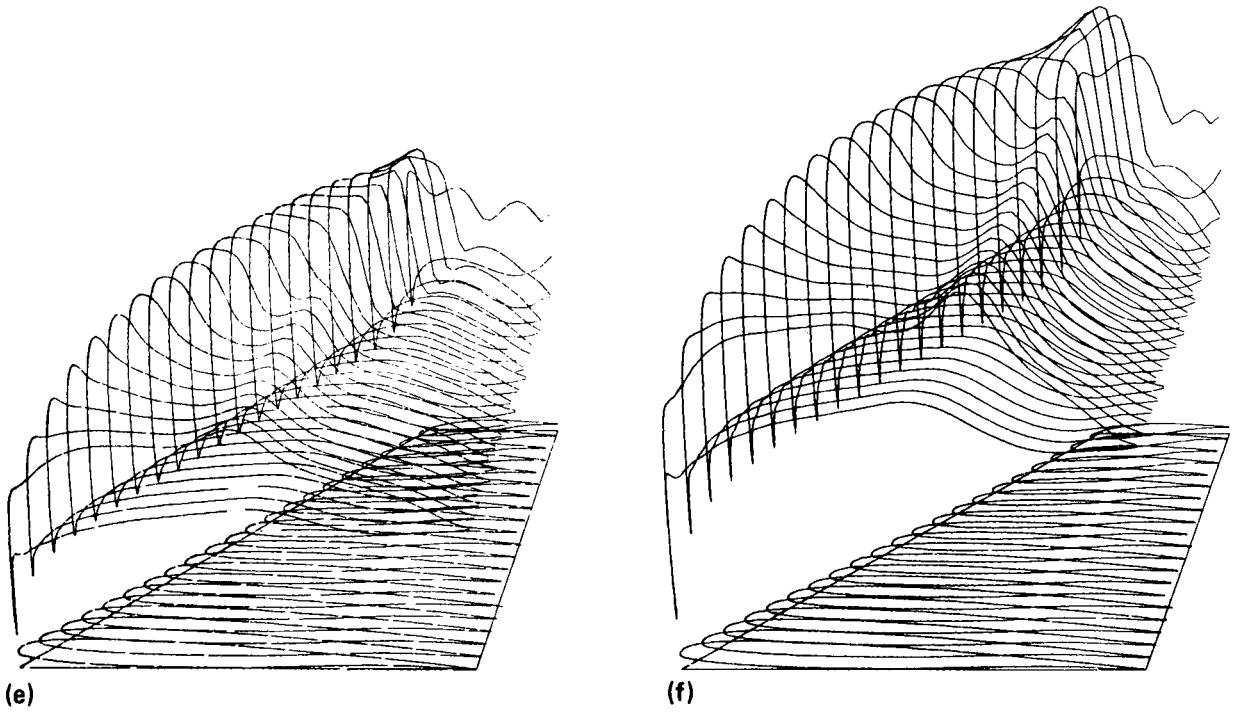
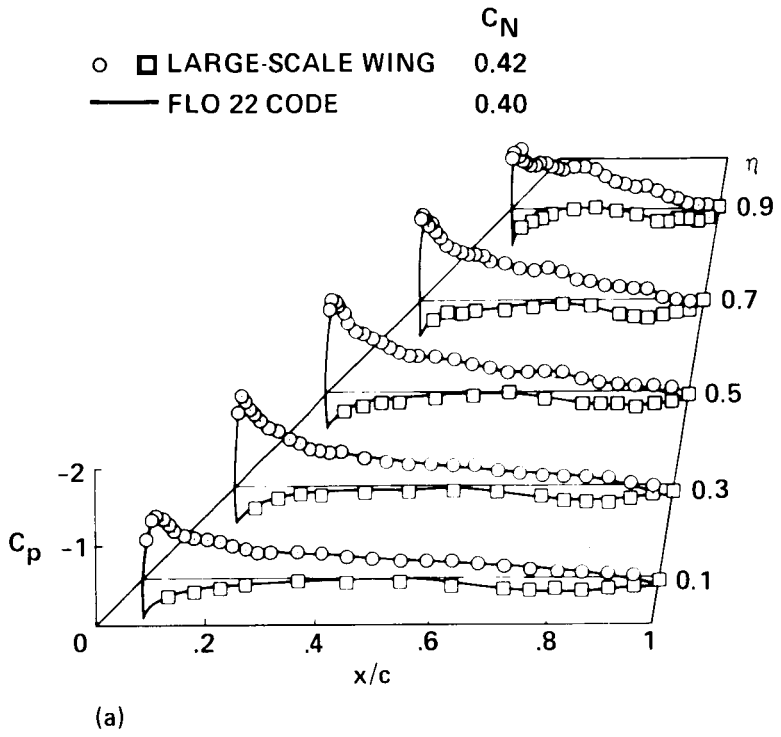
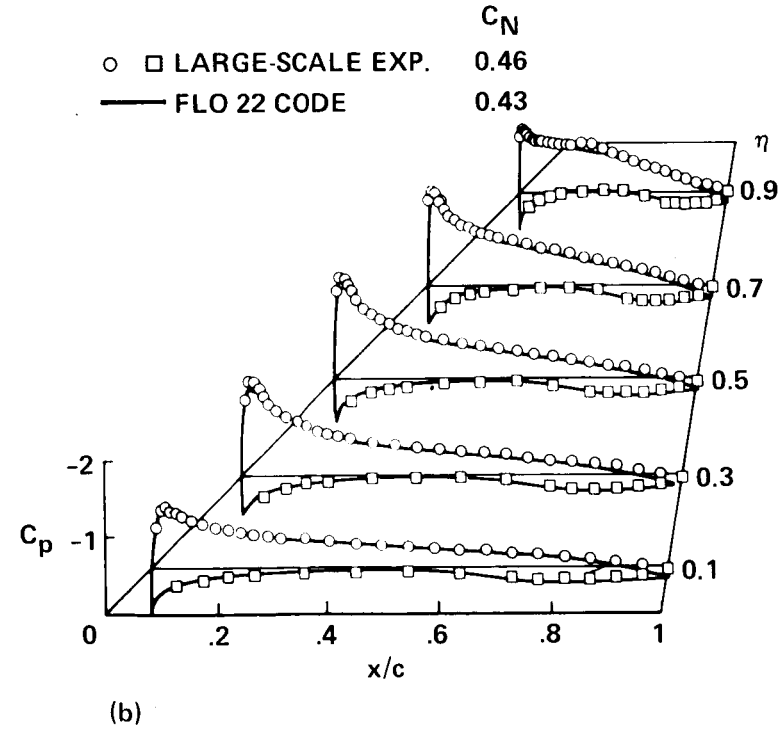


Figure 23.- Concluded. (e) M = 0.85 (a = 7°). (f) M = 0.90.

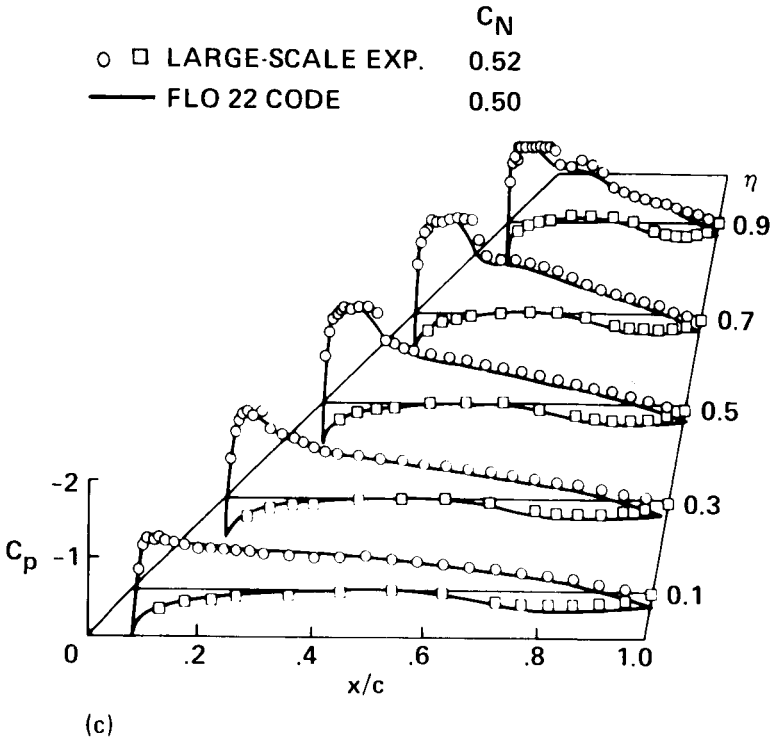


(a) M = 0.25.



(b) M = 0.50.

Figure 24.- Comparison of experimental and predicted pressure distributions by FL022 code at off-design conditions, a = 5°.



(c) M = 0.82.

Figure 24.- Concluded.

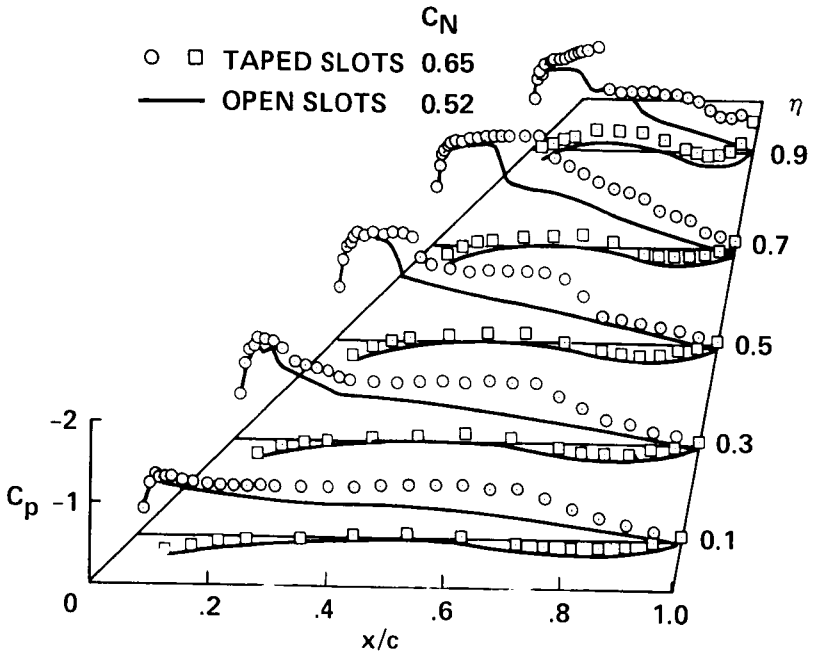


Figure 25.- Comparison of experimental pressure distributions with floor and ceiling suction slots open and taped to simulate solid walls; M = 0.82, a = 5°, Re = 6.8×10^6.

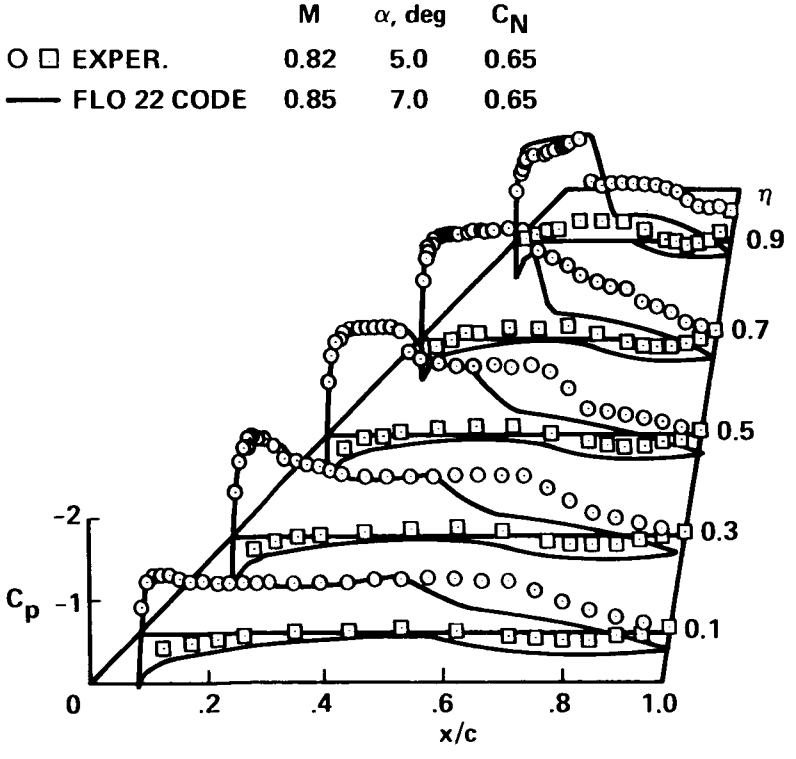
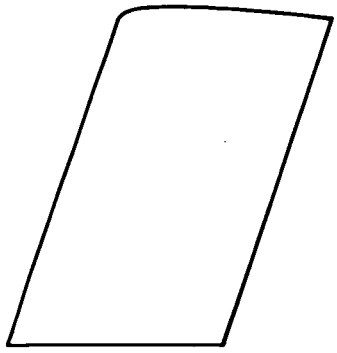
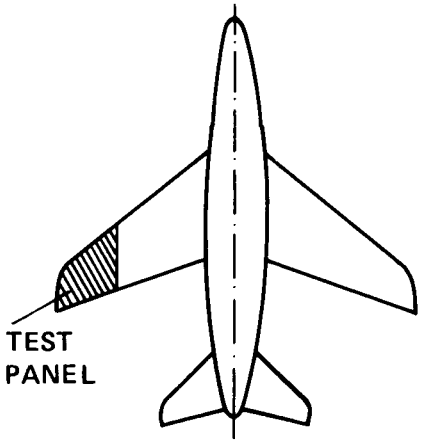


Figure 26.- Comparison of experimental pressure distributions with suction slots taped at M = 0.82 and a = 5° to those predicted by free-air code FL022 at M = 0.85 and a = 7°.



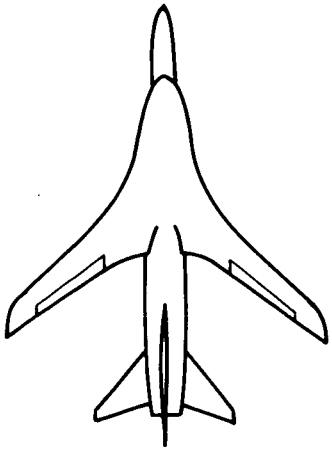
SWEPT NACA 0012
AIRFOIL (REF. 7)

ASPECT RATIO	3.0
L. E. SWEEP	20°
TAPER RATIO	0



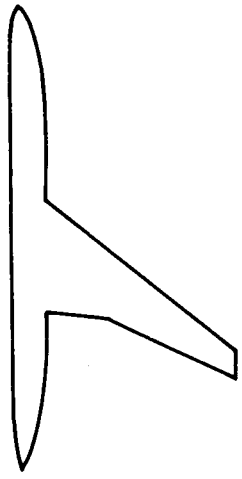
SAAB 32 LANSEN (REF. 10)
NACA 64 A010 AIRFOIL

	1.88
	39°
	0.61



F-8 SUPERCRITICAL WING
(REF. 9)

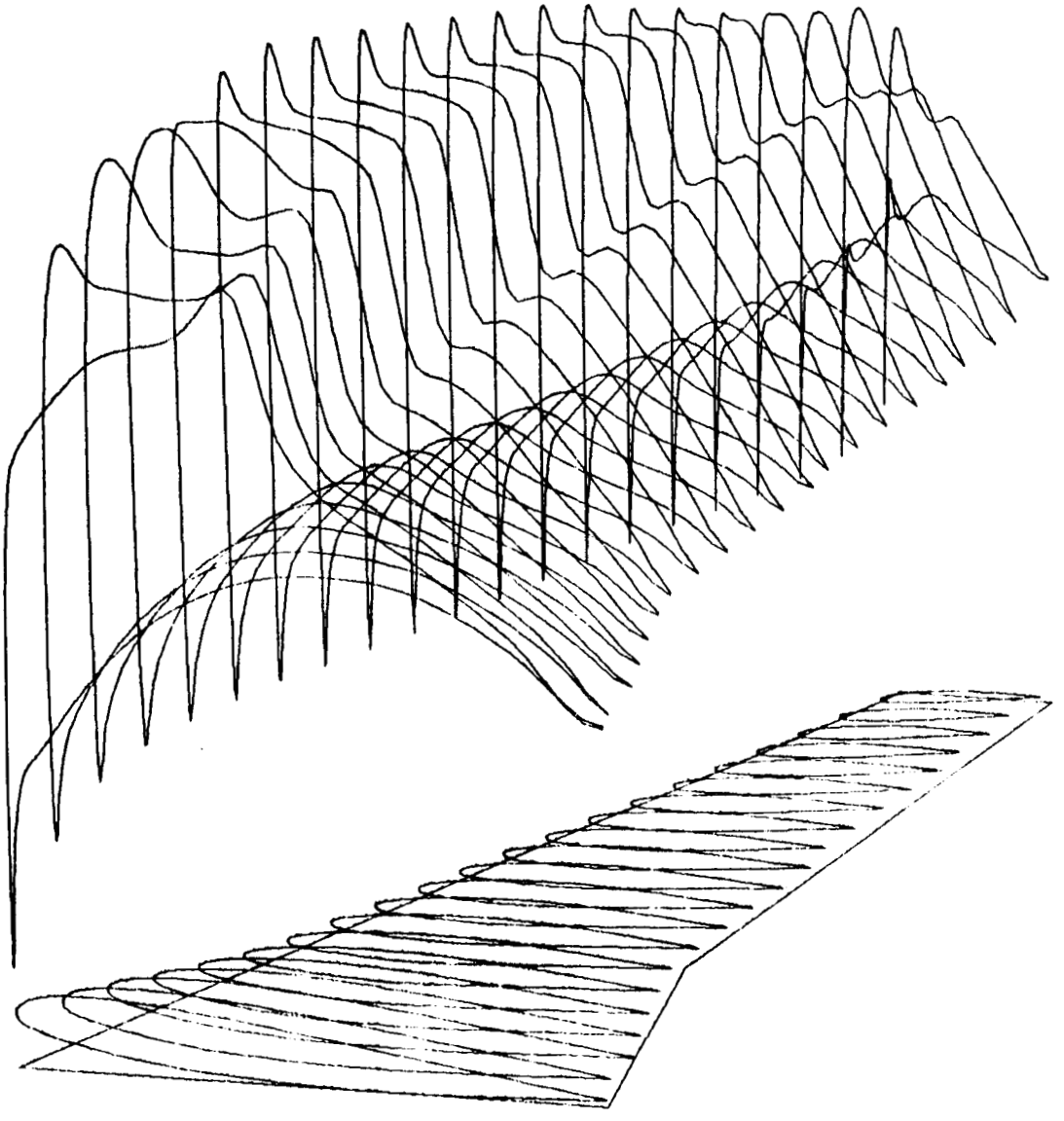
	6.77
	44°
	0.366



TRANSPORT MODEL
(REF. 11)

	6.8
	38°
	0.3

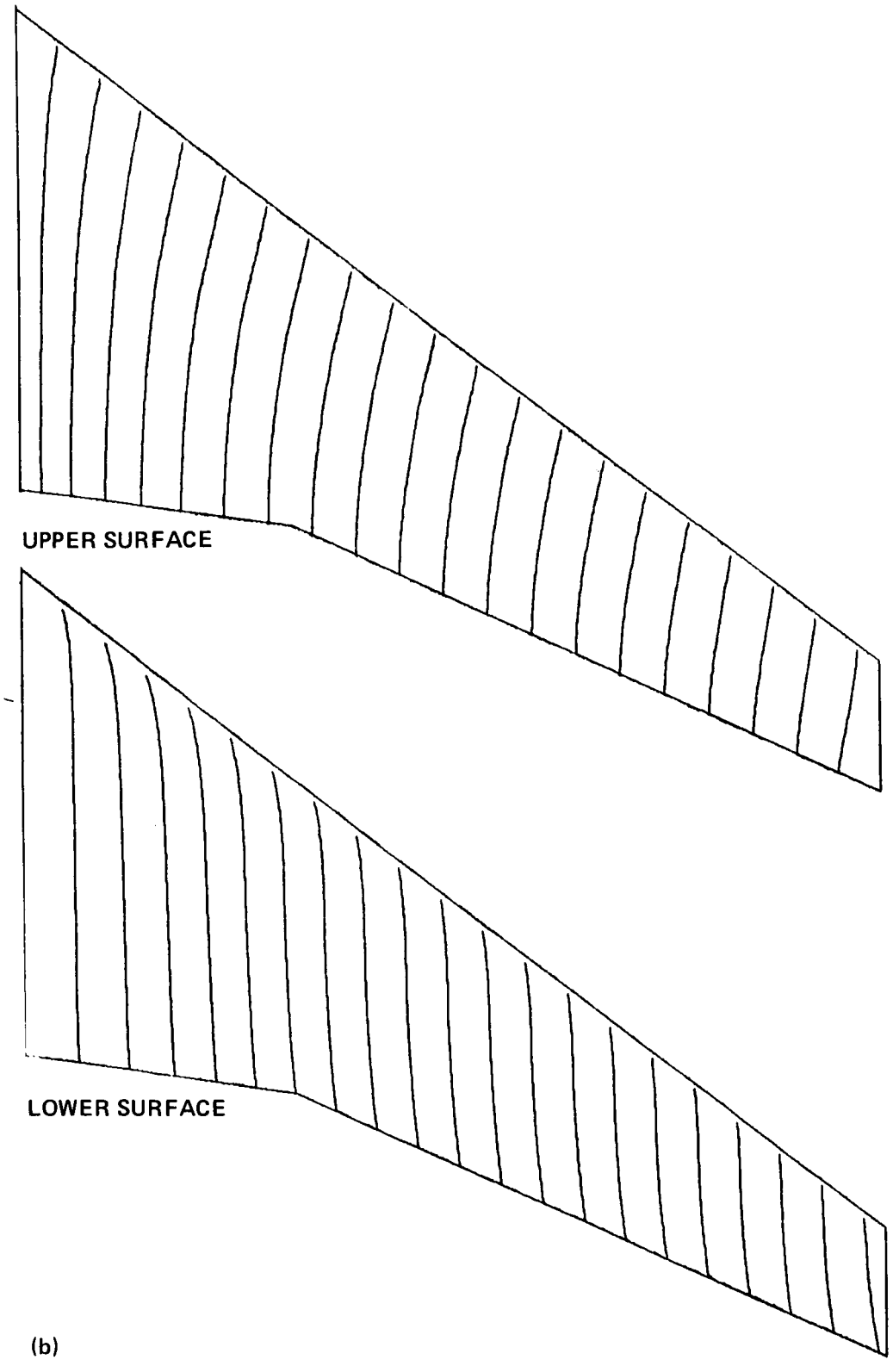
Figure 27.- Planform views of four wings whose data are relevant to the present results.



(a)

(a) Carpet of chordwise pressure distributions.

Figure 28.- Predicted inviscid wing chordwise pressure distributions by FL022 code for McDonnel-Douglas transport wing model; M = 0.85, a = 4°.



(b) Inviscid surface streamlines.

Figure 28.- Concluded.

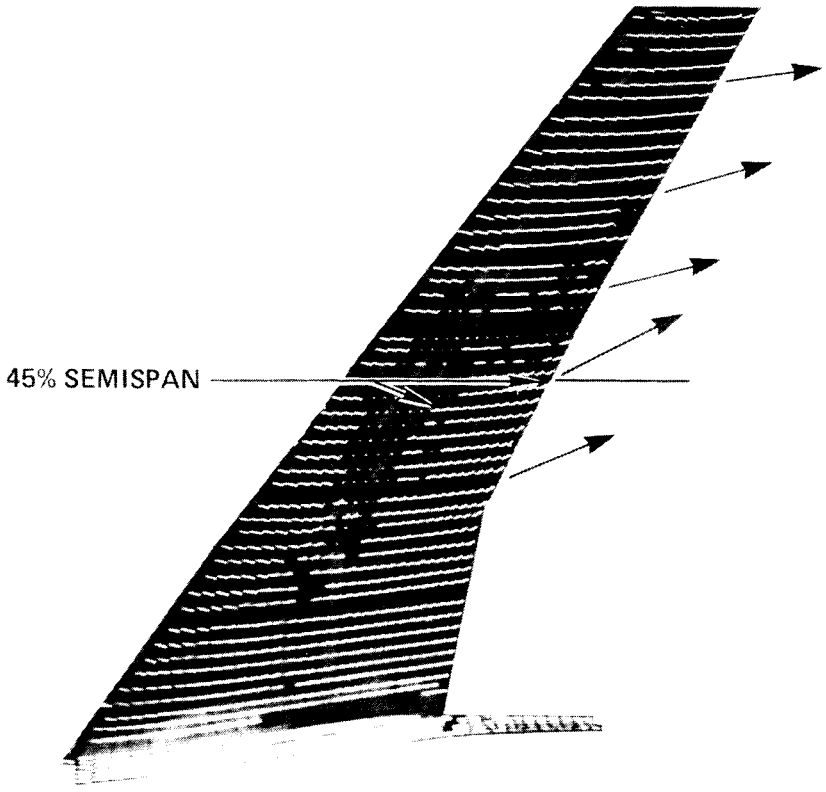


Figure 29.- Fluorescent mini-tuft flow visualization photograph of McDonnell-Douglas transport wing model (from ref. 5); M = 0.825, a = 4°.

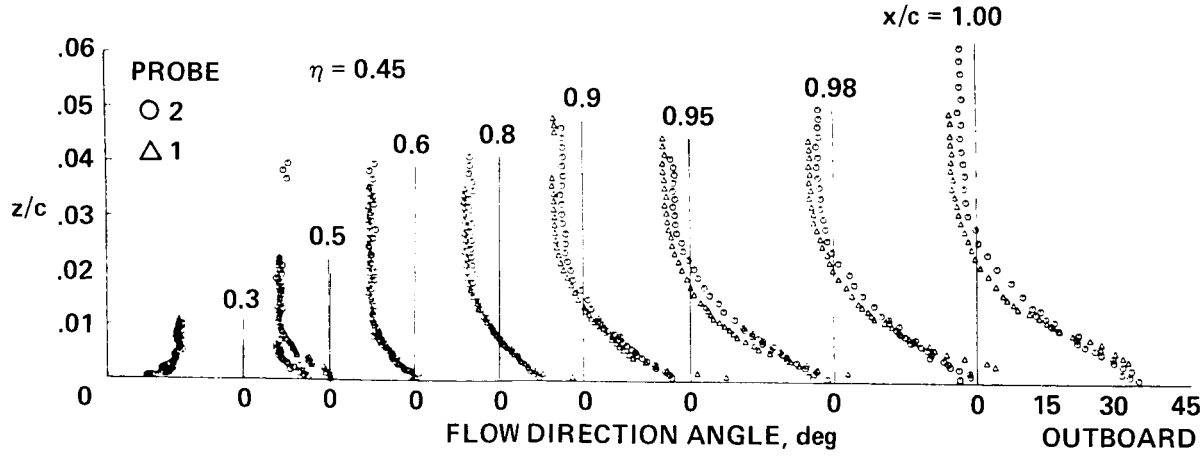


Figure 30. Boundary-layer flow-direction measurements over McDonnell-Douglas transport wing model (from ref. 5); M = 0.825, a = 4°.

START BSN 0860 WTTFAR** 21.41.53 02/18/84 DSNAME=S356.P1.T66.TRAN.T171802.PRPOUT0 FRICK STOP 10

ZDKCEXC BSN 0860

* PROGRAMMER.
* BUILDING.
* ROOM.
* DEPARTMENT.

```
 SSSSSSSSSS    TTTTTTTTTTTT    OOOOOOOOOO    PPPPPPPPPPPP                11          0000000 00
SSSSSSSSSSSS   TTTTTTTTTTTT   OOOOOOOOOOOO   PPPPPPPPPPPPP              111         0000000000
SS        SS       TT       OO        OO    PP          PP           1111        00      000
SS                 TT       OO        OO    PP          PP             11       00        00 00
SS                 TT       OO        OO    PP          PP             11       00       00  00
SSSSSSSSSSS        TT       OO        OO    PPPPPPPPPPPPP             11       00     00    00
 SSSSSSSSSSS       TT       OO        OO    PPPPPPPPPPPP              11       00   00      00
          SS       TT       OO        OO    PP                        11       00  00       00
          SS       TT       OO        OO    PP                        11       00 00        00
SS        SS       TT       OO        OO    PP                        11        000         00
SSSSSSSSSSSS       TT       OOOOOOOOOOOO    PP                  1111111111111    0000000000
 SSSSSSSSSS        TT        OOOOOOOOOO     PP                  1111111111111   00 0000000
```

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WW          WW   TTTTTTTTTTTT    TTTTTTTTTTTT    FFFFFFFFFFFFF      AAAAAAAAA      RRRRRRRRRRRR
WW          WW   TTTTTTTTTTTT    TTTTTTTTTTTT    FFFFFFFFFFFFF     AAAAAAAAAAA     RRRRRRRRRRRRR
WW          WW        TT             TT         FF               AA         AA     RR          RR
WW          WW        TT             TT         FF               AA         AA     RR          RR
WW          WW        TT             TT         FF               AA         AA     RR          RR
WW     W    WW        TT             TT         FFFFFFFFF        AA         AA     RRRRRRRRRRRRR
WW    WWW   WW        TT             TT         FFFFFFFFF        AAAAAAAAAAAAA     RRRRRRRRRRRR
WW   WW WW  WW        TT             TT         FF               AAAAAAAAAAAAA     RR       RR
WW WW   WW WW         TT             TT         FF               AA         AA     RR        RR
WWW      WWW          TT             TT         FF               AA         AA     RR         RR
WW        WW          TT             TT         FF               AA         AA     RR          RR
```

START BSN 0860 WTTFAR** 21.41.53 02/18/84 DSNAME=S356.P1.T66.TRAN.T171802.PRPOUT0 FRICK STOP 10

ZDKCEXC BSN 0860

* PROGRAMMER.
* BUILDING.
* ROOM.
* DEPARTMENT.

```
 SSSSSSSSSS    TTTTTTTTTTTT    00000  0000    PPPPPPPPPPP
SSSSSSSSSSSS   TTTTTTTTTTTT   000000  000000  PPPPPPPPPPPP
SS        SS       TT        00      00      PP        PP
SS                 TT        00      00      PP        PP
SS                 TT        00      00      PP        PP
SSSSSSSSSSS        TT        00      00      PPPPPPPPPPPP
 SSSSSSSSSSS       TT        00      00      PPPPPPPPPPP
          SS       TT        00      00      PP
          SS       TT        00      00      PP
SS        SS       TT        00      00      PP
SSSSSSSSSSSS       TT        0000000000000   PP
 SSSSSSSSSS        TT         00000000000    PP
```

```
    11          0000000 00
   111         0000000000
  1111        00      000
    11       00       00 00
    11       00      00  00
    11       00     00   00
    11       00    00    00
    11       00   00     00
    11       00  00      00
    11       00 00       00
    11        000       00
1111111111111  0000000000
1111111111111  00 0000000
```

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WW          WW  TTTTTTTTTTTT   TTTTTTTTTTTT   FFFFFFFFFFFFF     AAAAAAAAA      RRRRRRRRRRR
WW          WW  TTTTTTTTTTTT   TTTTTTTTTTTT   FFFFFFFFFFFFF    AAAAAAAAAAA     RRRRRRRRRRRR
WW          WW       TT            TT        FF              AA         AA     RR         RR
WW          WW       TT            TT        FF              AA         AA     RR         RR
WW          WW       TT            TT        FF              AA         AA     RR         RR
WW    W     WW       TT            TT        FFFFFFFFF       AA         AA     RRRRRRRRRRRR
WW   WWW    WW       TT            TT        FFFFFFFFF       AAAAAAAAAAAAA     RRRRRRRRRRR
WW  WW WW   WW       TT            TT        FF              AAAAAAAAAAAAA     RR       RR
WW WW   WW WW       TT            TT        FF              AA         AA     RR        RR
WWWW     WWWW       TT            TT        FF              AA         AA     RR         RR
WWW       WWW       TT            TT        FF              AA         AA     RR          RR
WW         WW       TT            TT        FF              AA         AA     RR          RR
```

F 260 T 269 TRANSMIT POUTO

RUN.SEQ
260.1

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.251	1 526	3.47	1911	1828	529.3	80.9	5.00	28

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.394	0.083	0.477	-0.0131	-0.0084	-0.0215	0.2025	28.32	35.11	29.51	42.41	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.326	0.087	0.413	-0.0317	-0.0156	-0.0473	34.72	42.84	36.44	0.592	0.152
0.296	0.393	0.090	0.483	-0.0239	-0.0171	-0.0410	31.08	43.92	33.48	0.590	0.016
0.500	0.443	0.097	0.540	-0.0268	-0.0185	-0.0454	31.06	44.12	33.40	0.540	-0.091
0.697	0.474	0.085	0.559	-0.0343	-0.0142	-0.0484	32.23	41.71	33.67	0.440	-0.123
0.894	0.463	0.036	0.499	-0.0502	-0.0074	-0.0576	35.85	45.41	36.54	0.286	-0.094

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.504	0.504	0.504	0.504	0.504
0.01			0.448		
0.02			0.450		
0.03			0.322		
0.04			0.301		
0.05	0.206	0.226	0.237	0.256	0.170
0.10	0.191	0.151	0.152	0.131	0.077
0.15	0.080	0.089	0.103	0.095	0.088
0.20	0.072	0.064	0.134	0.057	-0.011
0.30	0.033	0.035	0.021	0.081	-0.040
0.40	0.017	0.079	0.015	-0.009	-0.070
0.50	0.016	-0.009	-0.009	-0.007	-0.049
0.55					
0.60	0.097	0.039	0.051	0.024	-0.006
0.65			0.143		
0.70	0.078	0.096	0.102	0.119	0.134
0.75	0.113	0.123	0.140	0.136	0.085
0.80	0.109	0.136	0.128	0.202	0.107
0.85	0.106	0.187	0.141	0.126	0.089
0.90	0.083	0.087	0.105	0.113	0.088
0.95	0.105	0.059	0.127	0.030	0.044
1.00	-0.027	0.048	-0.016	-0.074	-0.068

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.195	-0.138	-0.338	-0.377	-0.111
0.003			-0.846		
0.006			-1.089	-1.054	
0.01	-0.602	-1.140	-1.270	-1.268	-0.890
0.02	-0.865	-1.311	-1.442	-1.328	-0.899
0.03	-0.862	-1.266	-1.365	-1.290	-0.935
0.04	-0.821	-1.168	-1.277	-1.162	-0.838
0.05	-0.793	-1.064	-1.173	-1.179	-0.917
0.06	-0.705	-0.985	-1.072	-1.085	-0.782
0.08	-0.593	-0.886	-0.914	-0.950	-0.764
0.10	-0.619	-0.815	-0.909	-0.884	-0.690
0.125	-0.556	-0.738	-0.816	-0.827	-0.728
0.15	-0.538	-0.676	-0.762	-0.727	-0.702
0.175	-0.504	-0.563	-0.709	-0.752	-0.702
0.20	-0.462	-0.584	-0.657	-0.704	-0.683
0.225	-0.382	-0.540	-0.565	-0.686	-0.670
0.25	-0.434	-0.523	-0.602	-0.650	-0.609
0.30	-0.394	-0.473	-0.543	-0.622	-0.684
0.35	-0.383	-0.423	-0.503	-0.532	-0.632
0.40	-0.356	-0.336	-0.462	-0.543	-0.608
0.45	-0.325	-0.379	-0.425	-0.490	-0.544
0.50	-0.251	-0.339	-0.328	-0.458	-0.486
0.55	-0.305	-0.325	-0.366	-0.400	-0.392
0.60	-0.269	-0.294	-0.324	-0.366	-0.412
0.65	-0.262	-0.257	-0.299	-0.264	-0.348
0.70	-0.237	-0.171	-0.261	-0.294	-0.312
0.75	-0.209	-0.215	-0.213	-0.243	-0.259
0.80	-0.121	-0.175	-0.134	-0.222	-0.224
0.85	-0.157	-0.155	-0.159	-0.159	-0.121
0.90	-0.097	-0.102	-0.097	-0.109	-0.153
0.95	-0.065	-0.045	-0.059	-0.002	-0.103
1.00	-0.027	0.048	-0.016	-0.074	-0.068

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR								
10.35	0.247	0.093			-0.328				
11.35	0.270	0.044			-0.234				
12.35	0.294				-0.268		-0.167		
14.35	0.342	0.078			-0.249				
15.35	0.366	0.085			-0.197				-0.120
16.35	0.390	0.126			-0.143				-5.545
17.35	0.413	0.177			-0.279		-0.247		-0.163
18.35	0.437	0.023			-0.191				-0.117
19.35	0.461	0.111			-0.231				-0 061
20.35	0.485	0.065			-0.214				-0.210
22.35	0.533	0.076			-0.152				-0.107
23.35	0.556	0.126			-0.094				-0.152
24.35	0.580	0.192			-0.230		-0.156		-0.141
25.35	0.604	0.029			-0.128				-0.096
26.35	0.628	0.131			-0.167				-0.035
27.35	0.652	0.098			-0.156				
30.35	0.723								-0.066
31.35	0.747		0.143	-0.115		-0.066	-0.059	-0.009	-0.100
32.35	0.771		0.193	-0.053		0.002		-0 146	-0.087
33.35	0.795		0.257	0.004		-0.150		-0.042	-0.035
34.35	0.818		0.094	-0.135		-0.042		-0.087	0.030
35.35	0.842		0.195	-0.033		-0.074		-0.077	-0.140
36.35	0.866		0.130	-0.057		-0.053		-0.016	-0.020
37.35	0.890		0.121	-0.027		0.019		0.049	-0.057
38.35	0.914		0.155	0.034		0.069		-0.097	-0.038
39.35	0.938		0.207	0.007	0.106	-0.065		0.003	0.008
40.35	0.961		0.036	0.029	-0.037	0.036		-0.021	0.062
41.35	0.958		0.116	0.101	0.079	0.001		-0.006	-0.093
42.35	1.009		0.083	0.166	0.042	0.018		0.049	-0.027
44.85	1.069		0.070	0.004	0.057	0.073		0.112	-0.023
45.85	1.092		0.120	0.090	0.090				
46.85	1.116		0.162	0.042	0.158				

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR							
16.75	0.399			-0.052	-0.176	-0.018		
13.75	0.328		-0.113	-0.074	0.013	-0.120		
10.75	0.256		-0.163	-0.141	-0.087	-0.020	-0.093	-0.027
7.75	0.185	-0.167	-0.247	-0.156	-0.059	0.036	-0.076	-0.010
6.75	0.161				-0.009	-0.016	-0.006	0.049 0.112
5.75	0.137	-0.244	-0.120	-0.182	-0.100	-0.014	0.093	0.045 -0.001
4.75	0.113				-0.115	-0.055	0.047	0.016 0.066
4.25	0.101	-0.268	-0.279	-0.230				
3.75	0.089				-0.066	-0.053	0.001	0.018 0.073
2.75	0.066				-0.083	-0.074	0.016	0.014 -0.041
1.75	0.042				-0.178	-0.012	0.022	0.075 0.151
0.75	0.018				-0.115	-0.057	0.079	0.042 0.057
-0.25	-0.006						0.101	0.166 0.004
-1.25	-0.030				0.143	0.130	0.116	0.083 0.070
-2.25	-0.054				0.172	0.134	0.195	0.195 0.080
-2.75	-0.066		0.177	0.192				
-3.25	-0.077				0.115	0.170	0.036	0.036 0.120
-4.25	-0.101	0.042	0.070	0.023	0.128	0.216	0.131	0.131 0.177
-5.25	-0.125				0.163	0.055	0.085	0.085 0.025
-6.25	-0.149	0.131	0.085	0.131	0.214	0.154	0.098	0.010
-9.25	-0.220		0.131	0.076	0.040	0.085	0.148	
-12.25	-0.292		0.184	0.087	0.129	0.096		
-15.25	-0.363			0.131	0.083	0.131		

RUN.SEQ
261.1

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.500	2.023	4.60	1384	1166	530.7	204.3	5.00	28

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.420	0.084	0.504	-0.0148	-0.0085	-0.0233	0.2144	28.53	35.09	29.63	42.57	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.347	0.087	0.435	-0.0343	-0.0134	-0.0477	34.89	40.35	35.99	0.623	0.163
0.296	0.417	0.090	0.507	-0.0270	-0.0172	-0.0442	31.48	44.16	33.72	0.618	0.015
0.500	0.467	0.098	0.565	-0.0280	-0.0199	-0.0479	31.00	45.20	33.47	0.565	-0.096
0.697	0.509	0.090	0.599	-0.0373	-0.0180	-0.0554	32.33	45.14	34.24	0.471	-0.134
0.894	0.500	0.031	0.532	-0.0536	-0.0085	-0.0621	35.71	51.98	36.67	0.305	-0.100

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.516	0.516	0.516	0.516	0.516
0.01			0.452		
0.02			0.394		
0.03			0.335		
0.04			0.305		
0.05	0.222	0.244	0.245	0.256	0.159
0.10	0.157	0.146	0.150	0.156	0.071
0.15	0.107	0.104	0.110	0.088	0.023
0.20	0.085	0.075	0.088	0.060	-0.009
0.30	0.055	0.026	0.040	0.024	-0.058
0.40	0.029	0.016	0.018	0.006	-0.065
0.50	0.018	-0.002	0.017	-0.008	-0.068
0.55					
0.60	0.045	0.047	0.052	0.054	-0.007
0.65			0.098		
0.70	0.093	0.128	0.125	0.128	0.090
0.75	0.114	0.141	0.145	0.139	0.113
0.80	0.131	0.149	0.160	0.160	0.115
0.85	0.114	0.145	0.157	0.149	0.120
0.90	0.088	0.107	0.122	0.126	0.090
0.95	0.055	0.073	0.086	0.084	0.048
1.00	-0.016	-0.003	0.003	-0.032	-0.044

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.160	-0.017	-0.187	-0.258	-0.060
0.003			-0.750		
0.006			-1.041	-1.035	
0.01	-0.670	-1.108	-1.319	-1.292	-0.900
0.02	-0.885	-1.341	-1.500	-1.405	-0.995
0.03	-0.885	-1.321	-1.439	-1.378	-0.989
0.04	-0.839	-1.231	-1.348	-1.293	-0.926
0.05	-0.800	-1.175	-1.238	-1.260	-0.971
0.06	-0.740	-1.030	-1.134	-1.167	-0.852
0.08	-0.675	-0.939	-1.019	-1.007	-0.811
0.10	-0.624	-0.856	-0.963	-0.948	-0.786
0.125	-0.579	-0.773	-0.863	-0.885	-0.766
0.15	-0.544	-0.711	-0.787	-0.829	-0.754
0.175	-0.519	-0.644	-0.723	-0.789	-0.732
0.20	-0.487	-0.611	-0.688	-0.751	-0.730
0.225	-0.468	-0.574	-0.642	-0.724	-0.735
0.25	-0.452	-0.532	-0.628	-0.700	-0.727
0.30	-0.429	-0.504	-0.569	-0.663	-0.739
0.35	-0.392	-0.443	-0.511	-0.618	-0.689
0.40	-0.375	-0.412	-0.476	-0.563	-0.640
0.45	-0.356	-0.392	-0.456	-0.531	-0.584
0.50	-0.321	-0.353	-0.411	-0.481	-0.515
0.55	-0.315	-0.322	-0.379	-0.441	-0.456
0.60	-0.293	-0.306	-0.333	-0.387	-0.416
0.65	-0.266	-0.279	-0.296	-0.336	-0.369
0.70	-0.256	-0.242	-0.275	-0.298	-0.310
0.75	-0.227	-0.228	-0.228	-0.250	-0.274
0.80	-0.186	-0.190	-0.188	-0.205	-0.242
0.85	-0.144	-0.142	-0.149	-0.165	-0.184
0.90	-0.103	-0.097	-0.099	-0.110	-0.144
0.95	-0.045	-0.049	-0.039	-0.053	-0.101
1.00	-0.016	-0.003	0.003	-0.032	-0.044

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR								
10.35	0.247	0.093			-0.328				
11.35	0.270	0.044			-0.234				
12.35	0.294				-0.268		-0.167		
14.35	0.342	0.078			-0.249				
15.35	0.366	0.085			-0.197				-0.120
16.35	0.390	0.126			-0.143				-5.545
17.35	0.413	0.177			-0.279		-0.247		-0.163
18.35	0.437	0.023			-0.191				-0.117
19.35	0.461	0.111			-0.231				-0.061
20.35	0.485	0.065			-0.214				-0.210
22.35	0.533	0.076			-0.152				-0.107
23.35	0.556	0.126			-0.094				-0.152
24.35	0.580	0.192			-0.230		-0.156		-0.141
25.35	0.604	0.029			-0.128				-0.096
26.35	0.628	0.131			-0.167				-0.035
27.35	0.652	0.098			-0.156				
30.35	0.723								-0.066
31.35	0.747		0.143	-0.115	-0.066	-0.059	-0.009	-0.100	
32.35	0.771		0.193	-0.053	0.002	-0.146	-0.087		
33.35	0.795		0.257	0.004	-0.150	-0.042	-0.035		
34.35	0.818		0.094	-0.135	-0.042	-0.087	0.030		
35.35	0.842		0.195	-0.033	-0.074	-0.077	-0.140		
36.35	0.866		0.130	-0.057	-0.053	-0.016	-0.020		
37.35	0.890		0.121	-0.027	0.019	0.049	-0.057		
38.35	0.914		0.155	0.034	0.069	-0.097	-0.038		
39.35	0.938		0.207	0.007	0.106	-0.065	0.003	0.008	
40.35	0.961		0.036	0.029	-0.037	0.036	-0.021	0.062	
41.35	0.958		0.116	0.101	0.079	0.001	-0.006	-0.093	
42.35	1.009		0.083	0.166	0.042	0.018	0.049	-0.027	
44.85	1.069		0.070	0.004	0.057	0.073	0.112	-0.023	
45.85	1.092		0.120	0.090	0.090				
46.85	1.116		0.162	0.042	0.158				

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR								
16.75	0.399			-0.052	-0.176	-0.018			
13.75	0.328		-0.113	-0.074	0.013	-0.120			
10.75	0.256		-0.163	-0.141	-0.087	-0.020	-0.093	-0.027	
7.75	0.185	-0.167	-0.247	-0.156	-0.059	0.036	-0.076		-0.010
6.75	0.161				-0.009	-0.016	-0.006	0.049	0.112
5.75	0.137	-0.244	-0.120	-0.182	-0.100	-0.014	0.093	0.045	-0.001
4.75	0.113				-0.115	-0.055	0.047	0.016	0.066
4.25	0.101	-0.268	-0.279	-0.230					
3.75	0.089				-0.066	-0.053	0.001	0.018	0.073
2.75	0.066				-0.083	-0.074	0.016	0.014	-0.041
1.75	0.042				-0.178	-0.012	0.022	0.075	0.151
0.75	0.018				-0.115	-0.057	0.079	0.042	0.057
-0.25	-0.006						0.101	0.166	0.004
-1.25	-0.030				0.143	0.130	0.116	0.083	0.070
-2.25	-0.054				0.172	0.134	0.195	0.195	0.080
-2.75	-0.066			0.177	0.192				
-3.25	-0.077					0.115	0.170	0.036	0.036	0.120
-4.25	-0.101	0.042	0.070	0.023	0.128	0.216	0.131	0.131	0.177
-5.25	-0.125				0.163	0.055	0.085	0.085	0.025
-6.25	-0.149	0.131	0.085	0.131	0.214	0.154	0.098		0.010
-9.25	-0.220		0.131	0.076	0.040	0.085	0.148		
-12.25	-0.292		0.184	0.087	0.129	0.096			
-15.25	-0.363			0.131	0.083	0.131			

RUN.SEQ
262.1

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.599	2.029	4.62	1217	955	532.1	239.9	5.00	28

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.429	0.088	0.518	-0.0156	-0.0094	-0.0250	0.2201	28.63	35.60	29.82	42.49	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.354	0.095	0.449	-0.0366	-0.0153	-0.0519	35.34	41.11	36.57	0.643	0.163
0.296	0.426	0.098	0.524	-0.0279	-0.0196	-0.0475	31.55	45.07	34.07	0.639	0.013
0.500	0.479	0.099	0.578	-0.0280	-0.0218	-0.0498	30.86	47.04	33.62	0.577	-0.099
0.697	0.520	0.091	0.611	-0.0349	-0.0199	-0.0549	31.72	46.83	33.98	0.481	-0.136
0.894	0.519	0.031	0.550	-0.0550	-0.0085	-0.0635	35.60	52.57	36.55	0.315	-0.103

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.523	0.523	0.523	0.523	0.523
0.01			0.454		
0.02			0.385		
0.03			0.328		
0.04			0.303		
0.05	0.226	0.247	0.245	0.248	0.150
0.10	0.160	0.144	0.154	0.149	0.075
0.15	0.112	0.100	0.096	0.085	0.031
0.20	0.096	0.069	0.082	0.057	-0.007
0.30	0.073	0.047	0.031	0.021	-0.067
0.40	0.046	0.038	0.013	-0.001	-0.062
0.50	0.006	0.017	0.004	0.004	-0.060
0.55					
0.60	0.045	0.044	0.059	0.056	-0.017
0.65			0.108		
0.70	0.111	0.128	0.136	0.132	0.079
0.75	0.129	0.144	0.161	0.152	0.108
0.80	0.131	0.158	0.164	0.166	0.126
0.85	0.124	0.157	0.157	0.158	0.122
0.90	0.106	0.119	0.143	0.133	0.093
0.95	0.067	0.081	0.095	0.088	0.056
1.00	-0.026	0.017	0.014	-0.015	-0.052

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.251	0.077	-0.100	-0.181	-0.029
0.003			-0.640		
0.006			-0.948	-0.960	
0.01	-0.588	-1.039	-1.256	-1.259	-0.918
0.02	-0.833	-1.351	-1.551	-1.457	-1.033
0.03	-0.860	-1.341	-1.495	-1.430	-1.017
0.04	-0.824	-1.267	-1.401	-1.345	-0.994
0.05	-0.794	-1.210	-1.363	-1.305	-1.032
0.06	-0.720	-1.048	-1.166	-1.260	-0.922
0.08	-0.671	-0.949	-1.042	-1.010	-0.855
0.10	-0.635	-0.860	-0.978	-0.986	-0.830
0.125	-0.588	-0.775	-0.888	-0.913	-0.807
0.15	-0.555	-0.723	-0.811	-0.864	-0.789
0.175	-0.516	-0.665	-0.748	-0.810	-0.764
0.20	-0.520	-0.634	-0.716	-0.774	-0.766
0.225	-0.493	-0.578	-0.669	-0.734	-0.749
0.25	-0.478	-0.546	-0.643	-0.715	-0.738
0.30	-0.437	-0.500	-0.587	-0.698	-0.747
0.35	-0.411	-0.478	-0.540	-0.656	-0.719
0.40	-0.386	-0.440	-0.503	-0.598	-0.665
0.45	-0.376	-0.411	-0.464	-0.534	-0.607
0.50	-0.337	-0.381	-0.427	-0.486	-0.540
0.55	-0.319	-0.344	-0.389	-0.441	-0.484
0.60	-0.305	-0.313	-0.346	-0.398	-0.438
0.65	-0.278	-0.290	-0.308	-0.345	-0.364
0.70	-0.256	-0.253	-0.277	-0.302	-0.309
0.75	-0.221	-0.225	-0.234	-0.242	-0.280
0.80	-0.185	-0.193	-0.202	-0.191	-0.248
0.85	-0.146	-0.150	-0.152	-0.148	-0.195
0.90	-0.106	-0.094	-0.085	-0.093	-0.150
0.95	-0.058	-0.033	-0.023	-0.034	-0.106
1.00	-0.026	0.017	0.014	-0.015	-0.052

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.093				-0.328				
11.35	0.270	0.044				-0.234				
12.35	0.294					-0.268		-0.167		
14.35	0.342	0.078				-0.249				
15.35	0.366	0.085				-0.197				-0.120
16.35	0.390	0.126				-0.143				-5.545
17.35	0.413	0.177				-0.279		-0.247		-0.163
18.35	0.437	0.023				-0.191				-0.117
19.35	0.461	0.111				-0.231				-0.061
20.35	0.485	0.065				-0.214				-0.210
22.35	0.533	0.076				-0.152				-0.107
23.35	0.556	0.126				-0.094				-0.152
24.35	0.580	0.192				-0.230		-0.156		-0.141
25.35	0.604	0.029				-0.128				-0.096
26.35	0.628	0.131				-0.167				-0.035
27.35	0.652	0.098				-0.156				
30.35	0.723									-0.066
31.35	0.747		0.143		-0.115		-0.066	-0.059	-0.009	-0.100
32.35	0.771		0.193		-0.053		0.002		-0.146	-0.087
33.35	0.795		0.257		0.004		-0.150		-0.042	-0.035
34.35	0.818		0.094		-0.135		-0.042		-0.087	0.030
35.35	0.842		0.195		-0.033		-0.074		-0.077	-0.140
36.35	0.866		0.130		-0.057		-0.053		-0.016	-0.020
37.35	0.890		0.121		-0.027		0.019		0.049	-0.057
38.35	0.914		0.155		0.034		0.069		-0.097	-0.038
39.35	0.938		0.207	0.007	0.106		-0.065		0.003	0.008
40.35	0.961		0.036	0.029	-0.037		0.036		-0.021	0.062
41.35	0.958		0.116	0.101	0.079		0.001		-0.006	-0.093
42.35	1.009		0.083	0.166	0.042		0.018		0.049	-0.027
44.85	1.069		0.070	0.004	0.057		0.073		0.112	-0.023
45.85	1.092		0.120	0.090	0.090					
46.85	1.116		0.162	0.042	0.158					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.58	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.052	-0.176	-0.018			
13.75	0.328		-0.113	-0.074	0.013	-0.120			
10.75	0.256		-0.163	-0.141	-0.087	-0.020	-0.093	-0.027	
7.75	0.185	-0.167	-0.247	-0.156	-0.059	0.036	-0.076		-0.010
6.75	0.161				-0.009	-0.016	-0.006	0.049	0.112
5.75	0.137	-0.244	-0.120	-0.182	-0.100	-0.014	0.093	0.045	-0.001
4.75	0.113				-0.115	-0.055	0.047	0.016	0.066
4.25	0.101	-0.268	-0.279	-0.230					
3.75	0.089				-0.066	-0.053	0.001	0.018	0.073
2.75	0.066				-0.083	-0.074	0.016	0.014	-0.041
1.75	0.042				-0.178	-0.012	0.022	0.075	0.151
0.75	0.018				-0.115	-0.057	0.079	0.042	0.057
-0.25	-0.006						0.101	0.166	0.004
-1.25	-0.030					0.143	0.130	0.116	0.083	0.070
-2.25	-0.054					0.172	0.134	0.195	0.195	0.080
-2.75	-0.066			0.177	0.192					
-3.25	-0.077					0.115	0.170	0.036	0.036	0.120
-4.25	-0.101	0.042	0.070	0.023	0.128	0.216	0.131	0.131	0.177
-5.25	-0.125					0.163	0.055	0.085	0.085	0.025
-6.25	-0.149	0.131	0.085	0.131	0.214	0.154	0.098		0.010
-9.25	-0.220		0.131	0.076	0.040	0.085	0.148		
-12.25	-0.292		0.184	0.087	0.129	0.096			
-15.25	-0.363			0.131	0.083	0.131			

RUN.SEQ
263.1

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.703	2.033	4.62	1102	793	533.9	274.2	5.00	28

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.471	0.074	0.545	-0.0197	-0.0083	-0.0280	0.2315	29.18	36.25	30.14	42.49	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.394	0.080	0.474	-0.0470	-0.0131	-0.0600	36.93	41.21	37.65	0.680	0.162
0.296	0.466	0.083	0.549	-0.0338	-0.0181	-0.0519	32.26	46.84	34.45	0.670	0.010
0.500	0.519	0.084	0.603	-0.0313	-0.0193	-0.0506	31.02	47.92	33.38	0.603	-0.102
0.697	0.570	0.073	0.643	-0.0374	-0.0176	-0.0550	31.56	49.12	33.55	0.506	-0.142
0.894	0.567	0.017	0.585	-0.0581	-0.0066	-0.0647	35.24	62.88	36.06	0.336	-0.109

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.532	0.532	0.532	0.532	0.532
0.01			0.438		
0.02			0.370		
0.03			0.319		
0.04			0.264		
0.05	0.204	0.224	0.207	0.218	0.127
0.10	0.137	0.131	0.126	0.121	0.052
0.15	0.102	0.091	0.087	0.069	0.000
0.20	0.086	0.062	0.063	0.024	-0.031
0.30	0.046	0.005	0.027	-0.006	-0.075
0.40	0.007	-0.005	-0.013	-0.024	-0.079
0.50	0.008	-0.018	-0.009	-0.026	-0.080
0.55					
0.60	0.023	0.042	0.037	0.039	-0.026
0.65			0.085		
0.70	0.101	0.132	0.121	0.130	0.074
0.75	0.115	0.147	0.162	0.146	0.106
0.80	0.127	0.169	0.171	0.156	0.114
0.85	0.127	0.154	0.161	0.151	0.118
0.90	0.092	0.117	0.123	0.124	0.088
0.95	0.054	0.072	0.080	0.086	0.047
1.00	-0.007	0.008	0.002	-0.022	-0.067

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.316	0.176	0.028	-0.058	0.010
0.003			-0.468		
0.006			-0.741	-0.784	
0.01	-0.500	-0.874	-1.051	-1.109	-0.933
0.02	-0.789	-1.277	-1.427	-1.409	-1.129
0.03	-0.841	-1.450	-1.604	-1.577	-1.133
0.04	-0.808	-1.385	-1.630	-1.613	-1.091
0.05	-0.792	-1.376	-1.697	-1.629	-1.155
0.06	-0.749	-1.185	-1.605	-1.558	-1.077
0.08	-0.702	-1.014	-1.028	-1.440	-0.915
0.10	-0.652	-0.949	-0.996	-0.995	-0.903
0.125	-0.621	-0.874	-0.940	-0.928	-0.893
0.15	-0.606	-0.782	-0.868	-0.927	-0.872
0.175	-0.579	-0.721	-0.807	-0.875	-0.860
0.20	-0.546	-0.671	-0.782	-0.849	-0.878
0.225	-0.524	-0.626	-0.741	-0.807	-0.846
0.25	-0.508	-0.592	-0.707	-0.784	-0.839
0.30	-0.500	-0.547	-0.630	-0.764	-0.848
0.35	-0.479	-0.529	-0.589	-0.727	-0.803
0.40	-0.462	-0.491	-0.550	-0.654	-0.726
0.45	-0.420	-0.467	-0.530	-0.593	-0.652
0.50	-0.401	-0.414	-0.476	-0.529	-0.564
0.55	-0.392	-0.388	-0.423	-0.475	-0.496
0.60	-0.358	-0.363	-0.367	-0.432	-0.446
0.65	-0.332	-0.327	-0.329	-0.377	-0.376
0.70	-0.299	-0.281	-0.296	-0.316	-0.325
0.75	-0.276	-0.252	-0.251	-0.257	-0.292
0.80	-0.245	-0.215	-0.205	-0.214	-0.267
0.85	-0.200	-0.166	-0.160	-0.166	-0.212
0.90	-0.109	-0.111	-0.101	-0.098	-0.172
0.95	-0.053	-0.044	-0.032	-0.035	-0.123
1.00	-0.007	0.008	0.002	-0.022	-0.067

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR								
10.35	0.247	0.093			-0.328				
11.35	0.270	0.044			-0.234				
12.35	0.294				-0.268		-0.167		
14.35	0.342	0.078			-0.249				
15.35	0.366	0.085			-0.197				-0.120
16.35	0.390	0.126			-0.143				-5.545
17.35	0.413	0.177			-0.279		-0.247		-0.163
18.35	0.437	0.023			-0.191				-0.117
19.35	0.461	0.111			-0.231				-0.061
20.35	0.485	0.065			-0.214				-0.210
22.35	0.533	0.076			-0.152				-0.107
23.35	0.556	0.126			-0.094				-0.152
24.35	0.580	0.192			-0.230		-0.156		-0.141
25.35	0.604	0.029			-0.128				-0.096
26.35	0.628	0.131			-0.167				-0.035
27.35	0.652	0.098			-0.156				
30.35	0.723								-0.066
31.35	0.747		0.143	-0.115	-0.066	-0.059	-0.009	-0.100	
32.35	0.771		0.193	-0.053	0.002		-0.146	-0.087	
33.35	0.795		0.257	0.004	-0.150		-0.042	-0.035	
34.35	0.818		0.094	-0.135	-0.042		-0.087	0.030	
35.35	0.842		0.195	-0.033	-0.074		-0.077	-0.140	
36.35	0.866		0.130	-0.057	-0.053		-0.016	-0.020	
37.35	0.890		0.121	-0.027	0.019		0.049	-0.057	
38.35	0.914		0.155	0.034	0.069		-0.097	-0.038	
39.35	0.938		0.207	0.007	0.106	-0.065	0.003	0.008	
40.35	0.961		0.036	0.029	-0.037	0.036	-0.021	0.062	
41.35	0.958		0.116	0.101	0.079	0.001	-0.006	-0.093	
42.35	1.009		0.083	0.166	0.042	0.018	0.049	-0.027	
44.85	1.069		0.070	0.004	0.057	0.073	0.112	-0.023	
45.85	1.092		0.120	0.090	0.090				
46.85	1.116		0.162	0.042	0.158				

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR								
16.75	0.399			-0.052	-0.176	-0.018			
13.75	0.328		-0.113	-0.074	0.013	-0.120			
10.75	0.256		-0.163	-0.141	-0.087	-0.020	-0.093	-0.027	
7.75	0.185	-0.167	-0.247	-0.156	-0.059	0.036	-0.076		-0.010
6.75	0.161				-0.009	-0.016	-0.006	0.049	0.112
5.75	0.137	-0.244	-0.120	-0.182	-0.100	-0.014	0.093	0.045	-0.001
4.75	0.113				-0.115	-0.055	0.047	0.016	0.066
4.25	0.101	-0.268	-0.279	-0.230					
3.75	0.089				-0.066	-0.053	0.001	0.018	0.073
2.75	0.066				-0.083	-0.074	0.016	0.014	-0.041
1.75	0.042				-0.178	-0.012	0.022	0.075	0.151
0.75	0.018				-0.115	-0.057	0.079	0.042	0.057
-0.25	-0.006						0.101	0.166	0.004
-1.25	-0.030				0.143	0.130	0.116	0.083	0.070
-2.25	-0.054				0.172	0.134	0.195	0.195	0.080
-2.75	-0.066		0.177	0.192					
-3.25	-0.077				0.115	0.170	0.036	0.036	0.120
-4.25	-0.101	0.042	0.070	0.023	0.128	0.216	0.131	0.131	0.177
-5.25	-0.125				0.163	0.055	0.085	0.085	0.025
-6.25	-0.149	0.131	0.085	0.131	0.214	0.154	0.098		0.010
-9.25	-0.220		0.131	0.076	0.040	0.085	0.148		
-12.25	-0.292		0.184	0.087	0.129	0.096			
-15.25	-0.363			0.131	0.083	0.131			

RUN.SEQ
264.1

MACH	RN/L	PN	PT	P	TTR	Q	ALPHA	CONF
0.499	3.029	5.89	2093	1766	534.2	307.9	5.00	28

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.420	0.085	0.505	-0.0150	-0.0088	-0.0238	0.2146	28.57	35.31	29.71	42.50	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.347	0.089	0.436	-0.0345	-0.0143	-0.0488	34.95	41.00	36.19	0.625	0.162
0.296	0.418	0.090	0.508	-0.0279	-0.0175	-0.0454	31.69	44.44	33.95	0.619	0.013
0.500	0.474	0.099	0.573	-0.0293	-0.0190	-0.0484	31.19	44.21	33.44	0.573	-0.097
0.697	0.505	0.090	0.594	-0.0360	-0.0182	-0.0543	32.14	45.31	34.13	0.468	-0.133
0.894	0.492	0.038	0.530	-0.0524	-0.0102	-0.0626	35.66	51.67	36.91	0.304	-0.100

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.516	0.516	0.516	0.516	0.516
0.01			0.453		
0.02			0.403		
0.03			0.333		
0.04			0.288		
0.05	0.227	0.236	0.293	0.247	0.157
0.10	0.164	0.148	0.154	0.150	0.078
0.15	0.113	0.092	0.098	0.092	0.042
0.20	0.077	0.074	0.086	0.060	0.001
0.30	0.054	0.031	0.032	0.037	-0.056
0.40	0.033	0.028	0.005	0.009	-0.053
0.50	0.012	0.002	0.001	-0.008	-0.055
0.55					
0.60	0.045	0.036	0.061	0.038	-0.001
0.65			0.115		
0.70	0.101	0.114	0.129	0.126	0.099
0.75	0.118	0.139	0.146	0.155	0.109
0.80	0.129	0.150	0.152	0.175	0.116
0.85	0.123	0.159	0.152	0.157	0.123
0.90	0.096	0.113	0.125	0.119	0.103
0.95	0.065	0.073	0.092	0.076	0.061
1.00	-0.026	0.005	-0.003	-0.022	-0.045

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.184	-0.010	-0.196	-0.273	-0.062
0.003			-0.748		
0.006			-1.047	-1.041	
0.01	-0.650	-1.116	-1.316	-1.289	-0.904
0.02	-0.878	-1.345	-1.502	-1.404	-0.976
0.03	-0.884	-1.321	-1.458	-1.369	-0.970
0.04	-0.809	-1.211	-1.341	-1.285	-0.937
0.05	-0.777	-1.084	-1.216	-1.243	-0.949
0.06	-0.746	-1.014	-1.153	-1.094	-0.828
0.08	-0.673	-0.943	-1.044	-1.017	-0.818
0.10	-0.629	-0.863	-0.972	-0.957	-0.779
0.125	-0.589	-0.785	-0.864	-0.875	-0.759
0.15	-0.560	-0.704	-0.791	-0.829	-0.756
0.175	-0.536	-0.641	-0.743	-0.792	-0.735
0.20	-0.486	-0.605	-0.703	-0.756	-0.730
0.225	-0.455	-0.574	-0.657	-0.721	-0.708
0.25	-0.448	-0.533	-0.635	-0.701	-0.694
0.30	-0.429	-0.485	-0.584	-0.653	-0.711
0.35	-0.394	-0.442	-0.531	-0.614	-0.681
0.40	-0.375	-0.407	-0.494	-0.558	-0.630
0.45	-0.351	-0.392	-0.457	-0.534	-0.576
0.50	-0.319	-0.380	-0.404	-0.481	-0.501
0.55	-0.308	-0.345	-0.379	-0.427	-0.441
0.60	-0.301	-0.313	-0.349	-0.385	-0.402
0.65	-0.270	-0.267	-0.312	-0.328	-0.370
0.70	-0.245	-0.229	-0.277	-0.294	-0.317
0.75	-0.219	-0.220	-0.222	-0.256	-0.271
0.80	-0.170	-0.201	-0.180	-0.212	-0.218
0.85	-0.153	-0.156	-0.153	-0.162	-0.171
0.90	-0.106	-0.102	-0.110	-0.097	-0.146
0.95	-0.057	-0.049	-0.050	-0.034	-0.109
1.00	-0.026	0.005	-0.008	-0.022	-0.045

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.093				-0.328				
11.35	0.270	0.044				-0.234				
12.35	0.294					-0.268		-0.167		
14.35	0.342	0.078				-0.249				
15.35	0.366	0.085				-0.197				-0.120
16.35	0.390	0.126				-0.143				-5.545
17.35	0.413	0.177				-0.279		-0.247		-0.163
18.35	0.437	0.023				-0.191				-0.117
19.35	0.461	0.111				-0.231				-0.061
20.35	0.485	0.065				-0.214				-0.210
22.35	0.533	0.076				-0.152				-0.107
23.35	0.556	0.126				-0.094				-0.152
24.35	0.580	0.192				-0.230		-0.156		-0.141
25.35	0.604	0.029				-0.128				-0.096
26.35	0.628	0.131				-0.167				-0.035
27.35	0.652	0.098				-0.156				
30.35	0.723									-0.066
31.35	0 747		0.143		-0.115		-0.066	-0.059	-0.009	-0.100
32.35	0.771		0.193		-0.053		0.002		-0.146	-0.087
33.35	0.795		0.257		0.004		-0.150		-0.042	-0.035
34.35	0.818		0.094		-0.135		-0.042		-0.087	0.030
35.35	0.842		0.195		-0.033		-0.074		-0.077	-0.140
36.35	0.866		0.130		-0.057		-0.053		-0.016	-0.020
37.35	0.890		0.121		-0.027		0.019		0.049	-0.057
38.35	0.914		0.155		0.034		0.069		-0.097	-0.038
39.35	0.938		0.207	0.007	0.106		-0.065		0.003	0.008
40.35	0.961		0.036	0.029	-0.037		0.036		-0.021	0.062
41.35	0.958		0.116	0.101	0.079		0.001		-0.006	-0.093
42.35	1.009		0.083	0.166	0.042		0.018		0.049	-0.027
44.85	1.069		0.070	0.004	0.057		0.073		0.112	-0.023
45.85	1.092		0.120	0.090	0.090					
46.85	1.116		0.162	0.042	0.158					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.052	-0.176	-0.018			
13.75	0.328		-0.113	-0.074	0.013	-0.120			
10.75	0.256		-0.163	-0.141	-0.087	-0.020	-0.093	-0.027	
7.75	0.185	-0.167	-0.247	-0.156	-0.059	0.036	-0.076		-0.010
6.75	0.161				-0.009	-0.016	-0.006	0.049	0.112
5.75	0.137	-0.244	-0.120	-0.182	-0.100	-0.014	0.093	0.045	-0.001
4.75	0.113				-0.115	-0.055	0.047	0.016	0.066
4.25	0.101	-0.268	-0.279	-0.2?0					
3.75	0.089				-0.066	-0.053	0.001	0.018	0.073
2.75	0.066				-0.083	-0.074	0.016	0.014	-0.041
1.75	0.042				-0.178	-0.012	0.022	0.075	0.151
0.75	0.018				-0.115	-0.057	0.079	0.042	0.057
-0.25	-0.006						0.101	0.166	0.004
-1.25	-0.030				0.143	0.130	0.116	0.083	0.070
-2.25	-0.054				0.172	0.134	0.195	0.195	0.080
-2.75	-0.066		0.177	0.192					
-3.25	-0.077				0.115	0.170	0.036	0.036	0.120
-4.25	-0.101	0.042	0.070	0.023	0.128	0.216	0.131	0.131	0.177
-5.25	-0.125				0.163	0.055	0.085	0.085	0.025
-6.25	-0.149	0.131	0.085	0.131	0.214	0.154	0.098		0.010
-9.25	-0.220		0.131	0.076	0.040	0.085	0.148		
-12.25	-0.29?		0.184	0.087	0.129	0.096			
-15.25	-0.363			0.131	0.083	0.131			

RUN.SEQ
265.1

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.498	3.009	6.85	2064	1742	530.1	302.0	5.00	28

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.406	0.092	0.498	-0.0136	-0.0091	-0.0227	0.2117	28.35	34.81	29.55	42.49	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.333	0.099	0.432	-0.0313	-0.0166	-0.0479	34.40	41.78	36.09	0.619	0.161
0 296	0.398	0.105	0.502	-0.0236	-0.0205	-0.0441	30.94	44.60	33.78	0.613	0.014
0.500	0.453	0.106	0.559	-0.0259	-0.0203	-0.0461	30.71	44.15	33.26	0.558	-0.093
0.697	0.497	0.090	0.587	-0.0349	-0.0186	-0.0535	32.02	45.69	34.11	0.462	-0.131
0.894	0.499	0.030	0.529	-0.0540	-0.0081	-0.0620	35.81	52.24	36.73	0.304	-0.100

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.516	0.516	0.516	0.516	0.516
0.01			0.457		
0.02			0.403		
0.03			0.337		
0.04			0.302		
0.05	0.228	0.253	0.294	0.242	0.154
0.10	0.175	0.164	0.156	0.147	0.064
0.15	0.118	0.107	0.112	0.093	0.038
0.20	0.099	0.087	0.099	0.060	-0.009
0.30	0.065	0.053	0.044	0.040	-0.056
0.40	0.041	0.048	0.023	-0.003	-0.072
0.50	0.023	0.016	0.011	0.005	-0.076
0.55					
0.60	0.061	0.056	0.062	0.038	-0.007
0.65			0.112		
0.70	0.111	0.124	0.126	0.130	0.097
0.75	0.125	0.157	0.156	0.146	0.108
0.80	0.136	0.162	0.160	0.172	0.107
0.85	0.133	0.173	0.162	0.147	0.106
0.90	0.101	0.122	0.121	0.126	0.094
0.95	0.069	0.072	0.095	0.082	0.062
1.00	-0.012	0.019	0.004	-0.018	-0.054

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.201	-0.019	-0.197	-0.275	-0.056
0.003			-0.741		
0.006			-1.014	-1.013	
0.01	-0.623	-1.067	-1.290	-1.272	-0.914
0.02	-0.872	-1.300	-1.476	-1.392	-1.005
0.03	-0.868	-1.275	-1.410	-1.361	-0.990
0.04	-0.803	-1.208	-1.308	-1.281	-0.923
0.05	-0.764	-1.062	-1.196	-1.229	-0.952
0.06	-0.728	-0.988	-1.100	-1.076	-0.834
0.08	-0.661	-0.910	-0.979	-0.996	-0.821
0.10	-0.621	-0.838	-0.921	-0.936	-0.788
0.125	-0.576	-0.762	-0.847	-0.870	-0.778
0.15	-0.540	-0.681	-0.780	-0.824	-0.753
0.175	-0.512	-0.610	-0.721	-0.782	-0.723
0.20	-0.477	-0.589	-0.675	-0.744	-0.717
0.225	-0.440	-0.549	-0.625	-0.701	-0.722
0.25	-0.436	-0.514	-0.608	-0.680	-0.717
0.30	-0.410	-0.472	-0.564	-0.652	-0.729
0.35	-0.380	-0.432	-0.506	-0.617	-0.683
0.40	-0.363	-0.391	-0.475	-0.570	-0.629
0.45	-0.333	-0.381	-0.437	-0.515	-0.578
0.50	-0.307	-0.346	-0.380	-0.463	-0.524
0.55	-0.294	-0.317	-0.355	-0.422	-0.461
0.60	-0.286	-0.290	-0.333	-0.386	-0.427
0.65	-0.255	-0.257	-0.298	-0.325	-0.371
0.70	-0.233	-0.219	-0.265	-0.290	-0.317
0.75	-0.199	-0.212	-0.208	-0.238	-0.277
0.80	-0.162	-0.176	-0.169	-0.206	-0.235
0.85	-0.144	-0.132	-0.141	-0.153	-0.175
0.90	-0.093	-0.087	-0.090	-0.100	-0.152
0.95	-0.043	-0.025	-0.032	-0.029	-0.101
1.00	-0.012	0.019	0.004	-0.018	-0.054

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.013	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.093				-0.328				
11.35	0.270	0.044				-0.234				
12.35	0.294					-0.268		-0.167		
14.35	0.342	0.078				-0 249				
15.35	0.366	0.085				-0.197				-0.120
16.35	0.390	0.126				-0.143				-5.545
17.35	0.413	0.177				-0.279		-0.247		-0.163
18.35	0.437	0.023				-0.191				-0.117
19.35	0.461	0.111				-0.231				-0.061
20.35	0.485	0.065				-0.214				-0.210
22.35	0.533	0.076				-0.152				-0.107
23.35	0.556	0.126				-0.094				-0.152
24.35	0.580	0.192				-0.230		-0.156		-0.141
25.35	0.604	0.029				-0.128				-0.096
26.35	0.628	0.131				-0.167				-0.035
27.35	0.652	0.098				-0.156				
30.35	0.723									-0.066
31.35	0.747		0.143		-0.115		-0.066	-0.059	-0.009	-0.100
32.35	0.771		0.193		-0.053		0.002		-0.146	-0.087
33.35	0.795		0.257		0.004		-0.150		-0.042	-0.035
34.35	0.818		0.094		-0.135		-0.042		-0.087	0.030
35.35	0.842		0.195		-0.033		-0.074		-0.077	-0.140
36.35	0.866		0.130		-0.057		-0.053		-0.016	-0.020
37.35	0.890		0.121		-0.027		0.019		0.049	-0.057
38.35	0.914		0.155		0.034		0.069		-0.097	-0.038
39.35	0.938		0.207	0.007	0.106		-0.065		0.003	0.008
40.35	0.961		0.036	0.029	-0.037		0.036		-0.021	0.062
41.35	0.958		0.116	0.101	0.079		0.001		-0.006	-0.093
42.35	1.009		0.083	0.166	0.042		0.018		0.049	-0.027
44.85	1.069		0.070	0.004	0.057		0.073		0.112	-0.023
45.85	1.092		0.120	0.090	0.090					
46.85	1.116		0.162	0.042	0.158					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.052	-0.176	-0.018			
13.75	0.328		-0.113	-0.074	0.013	-0.120			
10.75	0.256		-0.163	-0.141	-0.087	-0.020	-0.093	-0.027	
7.75	0.185	-0.167	-0.247	-0.156	-0.059	0.036	-0.076		-0.010
6.75	0.161				-0.009	-0.016	-0.006	0.049	0.112
5.75	0.137	-0.244	-0.120	-0.182	-0.100	-0.014	0.093	0.045	-0.001
4.75	0.113				-0.115	-0.055	0.04	0.016	0.066
4.25	0.101	-0.268	-0.279	-0.230					
3.75	0.089				-0.066	-0.053	0.001	0.018	0.073
2.75	0.066				-0.083	-0.074	0.016	0.014	-0.041
1.75	0.042				-0.178	-0.012	0.022	0.075	0.151
0.75	0.018				-0.115	-0.057	0.079	0.042	0.057
-0.25	-0.006						0.101	0.166	0.004
-1.25	-0.030				0.143	0.130	0.116	0.083	0.070
-2.25	-0.054				0.172	0.134	0.195	0.195	0.080
-2.75	-0.066		0.177	0.192					
-3.25	-0.077				0.115	0.170	0.036	0.036	0.120
-4.25	-0.101	0.042	0.070	0.023	0.128	0.216	0.131	0.131	0.177
-5.25	-0.125				0.163	0.055	0.085	0.085	0.025
-6.25	-0.149	0.131	0.085	0.131	0.214	0.154	0.098		0.010
-9.25	-0.220		0.131	0.076	0.040	0.085	0.148		
-12.25	-0.292		0.184	0.087	0.129	0.096			
-15.25	-0.363			0.131	0.083	0.131			

RUN.SEQ
266.1

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.602	3.042	6.92	1826	1429	534.0	362.8	5.00	28

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.430	0.085	0.515	-0.0161	-0.0091	-0.0252	0.2190	28.73	35.72	29.89	42.52	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.354	0.091	0.445	-0.0369	-0.0149	-0.0518	35.42	41.35	36.63	0.638	0.162
0.296	0.426	0.092	0.518	-0.0281	-0.0188	-0.0469	31.60	45.55	34.07	0.631	0.013
0.500	0.481	0.099	0.580	-0.0288	-0.0206	-0.0494	30.99	45.68	33.51	0.580	-0.099
0.697	0.521	0.086	0.607	-0.0366	-0.0187	-0.0553	32.02	46.84	34.12	0.478	-0.136
0.894	0.517	0.030	0.546	-0.0543	-0.0090	-0.0634	35.52	55.20	36.60	0.314	-0.103

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.523	0.523	0.523	0.523	0.523
0.01			0.452		
0.02			0.388		
0.03			0.329		
0.04			0.289		
0.05	0.226	0.247	0.288	0.239	0.138
0.10	0.158	0.143	0.147	0.144	0.078
0.15	0.114	0.097	0.101	0.080	0.026
0.20	0.090	0.072	0.075	0.054	-0.007
0.30	0.059	0.024	0.031	0.018	-0.065
0.40	0.028	0.013	0.005	-0.004	-0.072
0.50	0.008	-0.003	0.003	-0.013	-0.071
0.55					
0.60	0.038	0.046	0.057	0.043	-0.011
0.65			0.102		
0.70	0.106	0.134	0.128	0.128	0.085
0.75	0.125	0.143	0.166	0.151	0.108
0.80	0.144	0.163	0.173	0.163	0.123
0.85	0.137	0.162	0.160	0.153	0.126
0.90	0.096	0.120	0.134	0.128	0.102
0.95	0.060	0.076	0.091	0.087	0.058
1.00	-0.020	0.009	0.000	-0.006	-0.052

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.250	0.079	-0.095	-0.168	-0.012
0.003			-0.640		
0.006			-0.923	-0.965	
0.01	-0.580	-1.036	-1.242	-1.268	-0.904
0.02	-0.829	-1.359	-1.540	-1.476	-1.020
0.03	-0.865	-1.347	-1.524	-1.441	-1.022
0.04	-0.813	-1.243	-1.421	-1.369	-0.986
0.05	-0.789	-1.108	-1.288	-1.348	-1.002
0.06	-0.736	-1.035	-1.169	-1.127	-0.882
0.08	-0.677	-0.957	-1.040	-1.037	-0.847
0.10	-0.641	-0.871	-0.979	-0.982	-0.824
0.125	-0.590	-0.788	-0.887	-0.915	-0.800
0.15	-0.552	-0.731	-0.822	-0.863	-0.780
0.175	-0.533	-0.664	-0.766	-0.814	-0.760
0.20	-0.512	-0.622	-0.722	-0.784	-0.767
0.225	-0.478	-0.583	-0.677	-0.747	-0.752
0.25	-0.467	-0.555	-0.646	-0.724	-0.751
0.30	-0.439	-0.512	-0.591	-0.690	-0.768
0.35	-0.405	-0.472	-0.529	-0.644	-0.726
0.40	-0.387	-0.427	-0.495	-0.584	-0.665
0.45	-0.372	-0.403	-0.466	-0.539	-0.600
0.50	-0.344	-0.380	-0.426	-0.491	-0.536
0.55	-0.328	-0.342	-0.393	-0.448	-0.471
0.60	-0.305	-0.323	-0.357	-0.400	-0.426
0.65	-0.273	-0.283	-0.313	-0.346	-0.378
0.70	-0.255	-0.246	-0.279	-0.303	-0.317
0.75	-0.232	-0.220	-0.228	-0.259	-0.279
0.80	-0.192	-0.195	-0.192	-0.208	-0.234
0.85	-0.157	-0.147	-0.146	-0.158	-0.178
0.90	-0.104	-0.102	-0.098	-0.099	-0.139
0.95	-0.051	-0.037	-0.034	-0.038	-0.111
1.00	-0.020	0.009	0.000	-0.006	-0.052

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.093				-0.328				
11.35	0.270	0.044				-0.234				
12.35	0.294					-0.268		-0.167		
14.35	0.342	0.078				-0.249				
15.35	0.366	0.085				-0.197				-0.120
16.35	0.390	0.126				-0.143				-5.545
17.35	0.413	0.177				-0.279		-0.247		-0.163
18.35	0.437	0.023				-0.191				-0.117
19.35	0.461	0.111				-0.231				-0.061
20.35	0.485	0.065				-0.214				-0.210
22.35	0.533	0.076				-0.152				-0.107
23.35	0.556	0.126				-0.094				-0.152
24.35	0.580	0.192				-0.230		-0.156		-0.141
25.35	0.604	0.029				-0.128				-0.096
26.35	0.628	0.131				-0.167				-0.035
27.35	0.652	0.098				-0.156				
30.35	0.723									-0.066
31.35	0.747		0.143		-0.115		-0.066	-0.059	-0.009	-0.100
32.35	0.771		0.193		-0.053		0.002		-0.146	-0.087
33.35	0.795		0.257		0.004		-0.150		-0.042	-0.035
34.35	0.818		0.094		-0.135		-0.042		-0.087	0.030
35.35	0.842		0.195		-0.033		-0.074		-0.077	-0.140
36.35	0.866		0.130		-0.057		-0.053		-0.016	-0.020
37.35	0.890		0.121		-0.027		0.019		0.049	-0.057
38.35	0.914		0.155		0.034		0.069		-0.097	-0.033
39.35	0.938		0.207	0.007	0.106		-0.065		0.003	0.008
40.35	0.961		0.036	0.029	-0.037		0.036		-0.021	0.062
41.35	0.958		0.116	0.101	0.079		0.001		-0.006	-0.093
42.35	1.009		0.083	0.166	0.042		0.018		0.049	-0.027
44.85	1.069		0.070	0.004	0.057		0.073		0.112	-0.023
45.85	1.092		0.120	0.090	0.090					
46.85	1.116		0.162	0.042	0.158					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.052	-0.176	-0.018			
13.75	0.328		-0.113	-0.074	0.013	-0.120			
10.75	0.256		-0.163	-0.141	-0.087	-0.020	-0.093	-0.027	
7.75	0.185	-0.167	-0.247	-0.156	-0.059	0.036	-0.076		-0.010
6.75	0.161				-0.009	-0.016	-0.006	0.049	0.112
5.75	0.137	-0.244	-0.120	-0.182	-0.100	-0.014	0.093	0.045	-0.001
4.75	0.113				-0.115	-0.055	0.047	0.016	0.066
4.25	0.101	-0.268	-0.279	-0.230					
3.75	0.089				-0.066	-0.053	0.001	0.018	0.073
2.75	0.066				-0.083	-0.074	0.016	0.014	-0.041
1.75	0.042				-0.178	-0.012	0.022	0.075	0.151
0.75	0.018				-0.115	-0.057	0.079	0.042	0.057
-0.25	-0.006						0.101	0.166	0.004
-1.25	-0.030				0.143	0.130	0.116	0.083	0.070
-2.25	-0.054				0.172	0.134	0.195	0.195	0.080
-2.75	-0.066		0.177	0.192					
-3.25	-0.077				0.115	0.170	0.036	0.036	0.120
-4.25	-0.101	0.042	0.070	0.023	0.128	0.216	0.131	0.131	0.177
-5.25	-0.125				0.163	0.055	0.085	0.085	0.025
-6.25	-0.149	0.131	0.085	0.131	0.214	0.154	0.098		0.010
-9.25	-0.220		0.131	0.076	0.040	0.085	0.148		
-12.25	-0.292		0.184	0.087	0.129	0.096			
-15.25	-0.363			0.131	0.083	0.131			

RUN.SEQ
267.1

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.698	3.027	6.89	1662	1200	537.2	409.6	5.00	28

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.476	0.070	0.546	-0.0204	-0.0077	-0.0281	0.2322	29.29	35.99	30.15	42.53	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.397	0.076	0.473	-0.0468	-0.0120	-0.0588	36.79	40.79	37.44	0.678	0.164
0.296	0.471	0.076	0.548	-0.0349	-0.0159	-0.0508	32.41	45.73	34.27	0.668	0.012
0.500	0.531	0.083	0.614	-0.0328	-0.0185	-0.0513	31.17	47.29	33.35	0.614	-0.103
0.697	0.571	0.073	0.643	-0.0400	-0.0179	-0.0579	32.00	49.70	34.00	0.506	-0.143
0.894	0.571	0.011	0.583	-0.0597	-0.0064	-0.0661	35.45	80.59	36.34	0.335	-0.109

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.531	0.531	0.531	0.531	0.531
0.01			0.431		
0.02			0.354		
0.03			0.304		
0.04			0.267		
0.05	0.214	0.219	0.261	0.217	0.122
0.10	0.136	0.123	0.127	0.119	0.045
0.15	0.096	0.076	0.074	0.057	-0.008
0.20	0.070	0.055	0.048	0.029	-0.040
0.30	0.040	0.011	0.007	-0.008	-0.087
0.40	0.011	-0.004	-0.011	-0.026	-0.096
0.50	-0.020	-0.017	-0.016	-0.027	-0.089
0.55					
0.60	0.025	0.025	0.039	0.034	-0.036
0.65			0.088		
0.70	0.099	0.113	0.130	0.124	0.071
0.75	0.122	0.140	0.150	0.151	0.104
0.80	0.131	0.153	0.165	0.159	0.114
0.85	0.120	0.152	0.161	0.152	0.123
0.90	0.093	0.114	0.124	0.124	0.098
0.95	0.050	0.064	0.081	0.089	0.050
1.00	-0.027	-0.004	-0.002	-0.003	-0.064

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.320	0.170	0.027	-0.059	0.024
0.003			-0.480		
0.006			-0.746	-0.796	
0.01	-0.502	-0.885	-1.061	-1.128	-0.916
0.02	-0.803	-1.288	-1.446	-1.426	-1.106
0.03	-0.867	-1.462	-1.623	-1.578	-1.124
0.04	-0.824	-1.371	-1.637	-1.606	-1.077
0.05	-0.801	-1.392	-1.711	-1.664	-1.208
0.06	-0.764	-1.136	-1.691	-1.655	-0.969
0.08	-0.711	-1.047	-1.087	-1.045	-0.942
0.10	-0.676	-0.951	-1.038	-0.989	-0.925
0.125	-0.632	-0.873	-0.956	-0.959	-0.903
0.15	-0.614	-0.793	-0.887	-0.926	-0.891
0.175	-0.583	-0.730	-0.820	-0.882	-0.870
0.20	-0.570	-0.691	-0.783	-0.847	-0.867
0.225	-0.544	-0.643	-0.733	-0.811	-0.851
0.25	-0.527	-0.607	-0.704	-0.790	-0.842
0.30	-0.505	-0.569	-0.651	-0.763	-0.864
0.35	-0.466	-0.513	-0.601	-0.724	-0.805
0.40	-0.448	-0.494	-0.567	-0.650	-0.733
0.45	-0.427	-0.462	-0.522	-0.601	-0.654
0.50	-0.405	-0.426	-0.469	-0.542	-0.574
0.55	-0.390	-0.394	-0.429	-0.490	-0.504
0.60	-0.346	-0.360	-0.389	-0.436	-0.452
0.65	-0.314	-0.328	-0.352	-0.376	-0.402
0.70	-0.295	-0.281	-0.310	-0.321	-0.334
0.75	-0.269	-0.258	-0.252	-0.276	-0.301
0.80	-0.233	-0.221	-0.221	-0.226	-0.261
0.85	-0.192	-0.169	-0.167	-0.172	-0.214
0.90	-0.128	-0.111	-0.102	-0.106	-0.172
0.95	-0.072	-0.055	-0.042	-0.045	-0.127
1.00	-0.027	-0.004	-0.002	-0.003	-0.064

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.093				-0.328				
11.35	0.270	0.044				-0.234				
12.35	0.294					-0.268		-0.167		
14.35	0.342	0.078				-0.249				
15.35	0.366	0.085				-0.197				-0.120
16.35	0.390	0.126				-0.143				-5.545
17.35	0.413	0.177				-0.279		-0.247		-0.163
18.35	0.437	0.023				-0.191				-0.117
19.35	0.461	0.111				-0.231				-0.061
20.35	0.485	0.065				-0.214				-0.210
22.35	0.533	0.076				-0.152				-0.107
23.35	0.556	0.126				-0.094				-0.152
24.35	0.580	0.192				-0.230		-0.156		-0.141
25.35	0.604	0.029				-0.128				-0.096
26.35	0.628	0.131				-0.167				-0.035
27.35	0.652	0.098				-0.156				
30.35	0.723									-0.066
31.35	0.747		0.143		-0.115		-0.066	-0.059	-0.009	-0.100
32.35	0.771		0.193		-0.053		0.002		-0.146	-0.087
33.35	0.795		0.257		0.004		-0.150		-0.042	-0.035
34.35	0.818		0.094		-0.135		-0.042		-0.087	0.030
35.35	0.842		0.195		-0.033		-0.074		-0.077	-0.140
36.35	0.866		0.130		-0.057		-0.053		-0.016	-0.020
37.35	0.890		0.121		-0.027		0.019		0.049	-0.057
38.35	0.914		0.155		0.034		0.069		-0.097	-0.038
39.35	0.938		0.207	0.007	0.106		-0.065		0.003	0.008
40.35	0.961		0.036	0.029	-0.037		0.036		-0.021	0.062
41.35	0.958		0.116	0.101	0.079		0.001		-0.006	-0.093
42.35	1.009		0.083	0.166	0.042		0.018		0.049	-0.027
44.85	1.069		0.070	0.004	0.057		0.073		0.112	-0.023
45.85	1.092		0.120	0.090	0.090					
46.85	1.116		0.162	0.042	0.158					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.052	-0.176	-0.018			
13.75	0.328		-0.113	-0.074	0.013	-0.120			
10.75	0.256		-0.163	-0.141	-0.087	-0.020	-0.093	-0.027	
7.75	0.185	-0.167	-0.247	-0.156	-0.059	0.036	-0.076		-0.010
6.75	0.161				-0.009	-0.016	-0.006	0.049	0.112
5.75	0.137	-0.244	-0.120	-0.182	-0.100	-0.014	0.093	0.045	-0.001
4.75	0.113				-0.115	-0.055	0.047	0.016	0.066
4.25	0.101	-0.268	-0.279	-0.230					
3.75	0.089				-0.066	-0.053	0.001	0.018	0.073
2.75	0.066				-0.083	-0.074	0.016	0.014	-0.041
1.75	0.042				-0.178	-0.012	0.022	0.075	0.151
0.75	0.018				-0.115	-0.057	0.079	0.042	0.057
-0.25	-0.006						0.101	0.166	0.004
-1.25	-0.030				0.143	0.130	0.116	0.083	0.070
-2.25	-0.054				0.172	0.134	0.195	0.195	0.080
-2.75	-0.066		0.177	0.192					
-3.25	-0.077				0.115	0.170	0.036	0.036	0.120
-4.25	-0.101	0.042	0.070	0.023	0.128	0.216	0.131	0.131	0.177
-5.25	-0.125				0.163	0.055	0.085	0.085	0.025
-6.25	-0.149	0.131	0.085	0.131	0.214	0.154	0.098		0.010
-9.25	-0.220		0.131	0.076	0.040	0.085	0.148		
-12.25	-0.292		0.184	0.087	0.129	0.096			
-15.25	-0.363			0.131	0.083	0.131			

RUN.SEQ
268.1

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.821	1.999	4.55	1007	646	538.1	305.3	5.00	28

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.647	0.002	0.649	-0.0585	0.0047	-0.0538	0.2796	34.04	-244.2	33.30	43.08	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.528	0.021	0.549	-0.0927	0.0033	-0.0894	42.57	9.272	41.30	0.786	0.146
0.296	0.629	0.017	0.646	-0.0765	-0.0042	-0.0807	37.16	49.26	37.49	0.788	-0.017
0.500	0.715	0.007	0.722	-0.0668	-0.0059	-0.0728	34.35	108.9	35.09	0.721	-0.134
0.697	0.811	-0.008	0.803	-0.0793	-0.0026	-0.0819	34.77	-8.024	35.20	0.632	-0.185
0.894	0.801	-0.104	0.698	-0.1227	0.0211	-0.1016	40.31	45.33	39.56	0.400	-0.138

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.544	0.544	0.544	0.544	0.544
0.01			0.335		
0.02			0.257		
0.03			0.206		
0.04			0.175		
0.05	0.176	0.150	0.112	0.102	-0.008
0.10	0.092	0.067	0.037	0.004	-0.093
0.15	0.051	0.016	-0.002	-0.055	-0.139
0.20	0.025	-0.016	-0.037	-0.065	-0.166
0.30	-0.017	-0.069	-0.083	-0.101	-0.214
0.40	-0.053	-0.082	-0.098	-0.116	-0.225
0.50	-0.089	-0.113	-0.099	-0.128	-0.215
0.55					
0.60	-0.057	-0.032	-0.040	-0.046	-0.142
0.65			0.013		
0.70	0.034	0.078	0.060	0.062	-0.021
0.75	0.079	0.107	0.117	0.088	0.020
0.80	0.082	0.120	0.131	0.114	0.039
0.85	0.071	0.112	0.123	0.105	0.036
0.90	0.043	0.075	0.078	0.073	-0.005
0.95	-0.007	0.028	0.025	0.035	-0.080
1.00	-0.067	-0.044	-0.032	-0.073	-0.287

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.390	0.323	0.225	0.162	0.164
0.003			-0.186		
0.006			-0.409	-0.443	
0.01	-0.335	-0.550	-0.664	-0.695	-0.638
0.02	-0.634	-0.909	-0.989	-0.971	-0.866
0.03	-0.741	-1.097	-1.153	-1.117	-0.949
0.04	-0.729	-1.150	-1.202	-1.176	-0.936
0.05	-0.749	-1.238	-1.280	-1.238	-1.065
0.06	-0.704	-1.237	-1.326	-1.276	-1.070
0.08	-0.690	-1.209	-1.328	-1.277	-1.098
0.10	-0.661	-1.172	-1.338	-1.300	-1.123
0.125	-0.651	-0.949	-1.333	-1.318	-1.131
0.15	-0.633	-0.932	-1.333	-1.333	-1.146
0.175	-0.638	-0.900	-1.332	-1.330	-1.161
0.20	-0.629	-0.840	-1.316	-1.335	-1.190
0.225	-0.628	-0.798	-1.107	-1.339	-1.219
0.25	-0.626	-0.774	-0.954	-1.342	-1.236
0.30	-0.613	-0.747	-0.883	-1.349	-1.290
0.35	-0.617	-0.740	-0.865	-1.323	-1.304
0.40	-0.622	-0.747	-0.882	-1.221	-0.817
0.45	-0.647	-0.747	-0.883	-1.041	-0.788
0.50	-0.659	-0.743	-0.880	-0.991	-0.765
0.55	-0.665	-0.746	-0.860	-0.919	-0.742
0.60	-0.634	-0.725	-0.730	-0.597	-0.699
0.65	-0.607	-0.660	-0.568	-0.492	-0.625
0.70	-0.609	-0.473	-0.385	-0.377	-0.599
0.75	-0.464	-0.343	-0.264	-0.295	-0.568
0.80	-0.356	-0.273	-0.210	-0.255	-0.550
0.85	-0.273	-0.206	-0.179	-0.209	-0.422
0.90	-0.186	-0.147	-0.126	-0.158	-0.383
0.95	-0.110	-0.097	-0.063	-0.096	-0.317
1.00	-0.067	-0.044	-0.032	-0.073	-0.287

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR								
10.35	0.247	0.093			-0.328				
11.35	0.270	0.044			-0.234				
12.35	0.294				-0.268		-0.167		
14.35	0.342	0.078			-0.249				
15.35	0.366	0.085			-0.197				-0.120
16.35	0.390	0.126			-0.143				-5.545
17.35	0.413	0.177			-0.279		-0.247		-0.163
18.35	0.437	0.023			-0.191				-0.117
19.35	0.461	0.111			-0.231				-0.061
20.35	0.485	0.065			-0.214				-0.210
22.35	0.533	0.076			-0.152				-0.107
23.35	0.556	0.126			-0.094				-0.152
24.35	0.580	0.192			-0.230		-0.156		-0.141
25.35	0.604	0.029			-0.128				-0.096
26.35	0.628	0.131			-0.167				-0.035
27.35	0.652	0.098			-0.156				
30.35	0.723								-0.066
31.35	0.747		0.143	-0.115		-0.066	-0.059	-0.009	-0.100
32.35	0.771		0.193	-0.053		0.002		-0.146	-0.087
33.35	0.795		0.257	0.004		-0.150		-0.042	-0.035
34.35	0.818		0.094	-0.135		-0.042		-0.087	0.030
35.35	0.842		0.195	-0.033		-0.074		-0.077	-0.140
36.35	0.866		0.130	-0.057		-0.053		-0.016	-0.020
37.35	0.890		0.121	-0.027		0.019		0.049	-0.057
38.35	0.914		0.155	0.034		0.069		-0.097	-0.038
39.35	0.938		0.207	0.007	0.106	-0.065		0.003	0.008
40.35	0.961		0.036	0.029	-0.037	0.036		-0.021	0.062
41.35	0.958		0.116	0.101	0.079	0.001		-0.006	-0.093
42.35	1.009		0.083	0.166	0.042	0.018		0.049	-0.027
44.85	1.069		0.070	0.004	0.057	0.073		0.112	-0.023
45.85	1.092		0.120	0.090	0.090				
46.85	1.116		0.162	0.042	0.158				

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR								
16.75	0.399			-0.052	-0.176	-0.018			
13.75	0.328		-0.113	-0.074	0.013	-0.120			
10.75	0.256		-0.163	-0.141	-0.087	-0.020	-0.093	-0.027	
7.75	0.185	-0.167	-0.247	-0.156	-0.059	0.036	-0.076		-0.010
6.75	0.161				-0.009	-0.016	-0.006	0.049	0.112
5.75	0.137	-0.244	-0.120	-0.182	-0.100	-0.014	0.093	0.045	-0.001
4.75	0.113				-0.115	-0.055	0.047	0.016	0.066
4.25	0.101	-0.268	-0.279	-0.230					
3.75	0.089				-0.066	-0.053	0.001	0.018	0.073
2.75	0.066				-0.083	-0.074	0.016	0.014	-0.041
1.75	0.042				-0.178	-0.012	0.022	0.075	0.151
0.75	0.018				-0.115	-0.057	0.079	0.042	0.057
-0.25	-0.006						0.101	0.166	0.004
-1.25	-0.030				0.143	0.130	0.116	0.083	0.070
-2.25	-0.054				0.172	0.134	0.195	0.195	0.080
-2.75	-0.066		0.177	0.192					
-3.25	-0.077				0.115	0.170	0.036	0.036	0.120
-4.25	-0.101	0.042	0.070	0.023	0.128	0.216	0.131	0.131	0.177
-5.25	-0.125				0.163	0.055	0.085	0.085	0.025
-6.25	-0.149	0.131	0.085	0.131	0.214	0.154	0.098		0.010
-9.25	-0.220		0.131	0.076	0.040	0.085	0.148		
-12.25	-0.292		0.184	0.087	0.129	0.096			
-15.25	-0.363			0.131	0.083	0.131			

RUN.SEQ
269.1

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.821	3.010	6.85	1533	985	542.5	464.3	5.00	28

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.648	0.001	0.648	-0.0645	0.0072	-0.0573	0.2766	34.96	-951.1	33.84	42.66	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.528	0.032	0.559	-0.0955	0.0010	-0.0945	43.09	21.87	41.89	0.802	0.142
0.296	0.632	0.014	0.646	-0.0831	-0.0020	-0.0850	38.15	39.26	38.17	0.788	-0.024
0.500	0.711	0.006	0.717	-0.0717	-0.0041	-0.0758	35.08	95.23	35.57	0.717	-0.137
0.697	0.832	-0.021	0.811	-0.0992	0.0019	-0.0973	36.92	33.82	37.00	0.638	-0.196
0.894	0.763	-0.114	0.649	-0.1267	0.0250	-0.1017	41.60	46.87	40.67	0.373	-0.131

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.544	0.544	0.544	0.544	0.544
0.01			0.336		
0.02			0.258		
0.03			0.205		
0.04			0.178		
0.05	0.177	0.146	0.157	0.096	-0.035
0.10	0.108	0.059	0.036	0.002	-0.078
0.15	0.066	0.011	-0.007	-0.055	-0.129
0.20	0.033	-0.017	-0.034	-0.077	-0.159
0.30	-0.011	-0.055	-0.084	-0.112	-0.242
0.40	-0.039	-0.076	-0.115	-0.130	-0.247
0.50	-0.058	-0.095	-0.120	-0.142	-0.226
0.55					
0.60	-0.033	-0.060	-0.030	-0.069	-0.137
0.65			0.029		
0.70	0.049	0.059	0.070	0.047	-0.025
0.75	0.072	0.102	0.112	0.081	0.005
0.80	0.085	0.114	0.126	0.098	0.039
0.85	0.085	0.111	0.119	0.091	0.029
0.90	0.038	0.062	0.065	0.060	-0.009
0.95	-0.007	0.015	0.020	0.012	-0.113
1.00	-0.072	-0.049	-0.050	-0.097	-0.383

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.405	0.329	0.235	0.179	0.185
0.003			-0.172		
0.006			-0.396	-0.430	
0.01	-0.320	-0.544	-0.650	-0.682	-0.611
0.02	-0.627	-0.908	-0.980	-0.956	-0.836
0.03	-0.731	-1.093	-1.147	-1.105	-0.928
0.04	-0.710	-1.134	-1.193	-1.155	-0.898
0.05	-0.722	-1.230	-1.276	-1.228	-1.046
0.06	-0.708	-1.209	-1.312	-1.266	-1.043
0.08	-0.676	-1.181	-1.301	-1.268	-1.067
0.10	-0.652	-1.108	-1.320	-1.276	-1.084
0.125	-0.633	-0.933	-1.307	-1.291	-1.099
0.15	-0.622	-0.916	-1.326	-1.300	-1.112
0.175	-0.618	-0.870	-1.316	-1.314	-1.133
0.20	-0.621	-0.832	-1.261	-1.315	-1.164
0.225	-0.613	-0.788	-1.014	-1.321	-1.189
0.25	-0.613	-0.749	-0.934	-1.317	-1.206
0.30	-0.608	-0.723	-0.868	-1.333	-1.245
0.35	-0.611	-0.725	-0.851	-1.331	-0.736
0.40	-0.618	-0.727	-0.863	-1.067	-0.705
0.45	-0.635	-0.729	-0.862	-0.975	-0.710
0.50	-0.643	-0.722	-0.853	-0.881	-0.712
0.55	-0.655	-0.731	-0.850	-0.784	-0.707
0.60	-0.657	-0.729	-0.775	-0.696	-0.694
0.65	-0.625	-0.721	-0.568	-0.664	-0.688
0.70	-0.613	-0.578	-0.326	-0.593	-0.643
0.75	-0.510	-0.430	-0.295	-0.447	-0.607
0.80	-0.377	-0.280	-0.252	-0.388	-0.498
0.85	-0.269	-0.227	-0.208	-0.324	-0.427
0.90	-0.187	-0.158	-0.165	-0.196	-0.411
0.95	-0.119	-0.093	-0.099	-0.149	-0.439
1.00	-0.072	-0.049	-0.050	-0.097	-0.383

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR								
10.35	0.247	0.093			-0.328				
11.35	0.270	0.044			-0.234				
12.35	0.294				-0.268		-0.167		
14.35	0.342	0.078			-0.249				
15.35	0.366	0.085			-0.197				-0.120
16.35	0.390	0.126			-0.143				-5.545
17.35	0.413	0.177			-0.279		-0.247		-0.163
18.35	0.437	0.023			-0.191				-0.117
19.35	0.461	0.111			-0.231				-0.061
20.35	0.485	0.065			-0.214				-0.210
22.35	0.533	0.076			-0.152				-0.107
23.35	0.556	0.126			-0.094				-0.152
24.35	0.580	0.192			-0.230		-0.156		-0.141
25.35	0.604	0.029			-0.128				-0.096
26.35	0.628	0.131			-0.167				-0.035
27.35	0.652	0.098			-0.156				
30.35	0.723								-0.066
31.35	0.747		0.143		-0.115		-0.066	-0.059	-0.009 -0.100
32.35	0.771		0.193		-0.053		0.002		-0.146 -0.087
33.35	0.795		0.257		0.004		-0.150		-0.042 -0.035
34.35	0.818		0.094		-0.135		-0.042		-0.087 0.030
35.35	0.842		0.195		-0.033		-0.074		-0.077 -0.140
36.35	0.866		0.130		-0.057		-0.053		-0.016 -0.020
37.35	0.890		0.121		-0.027		0.019		0.049 -0.057
38.35	0.914		0.155		0.034		0.069		-0.097 -0.038
39.35	0.938		0.207	0.007	0.106		-0.065		0.003 0.008
40.35	0.961		0.036	0.029	-0.037		0.036		-0.021 0.062
41.35	0.958		0.116	0.101	0.079		0.001		-0.006 -0.093
42.35	1.009		0.083	0.166	0.042		0.018		0.049 -0.027
44.85	1.069		0.070	0.004	0.057		0.073		0.112 -0.023
45.85	1.092		0.120	0.090	0.090				
46.85	1.116		0.162	0.042	0.158				

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR								
16.75	0.399			-0.052	-0.176	-0.018			
13.75	0.328		-0.113	-0.074	0.013	-0.120			
10.75	0.256		-0.163	-0.141	-0.087	-0.020	-0.093	-0.027	
7.75	0.185	-0.167	-0.247	-0.156	-0.059	0.036	-0.076		-0.010
6.75	0.161				-0.009	-0.016	-0.006	0.049	0.112
5.75	0.137	-0.244	-0.120	-0.182	-0.100	-0.014	0.093	0.045	-0.001
4.75	0.113				-0.115	-0.055	0.047	0.016	0.066
4.25	0.101	-0.268	-0.279	-0.230					
3.75	0.089				-0.066	-0.053	0.001	0.018	0.073
2.75	0.066				-0.083	-0.074	0.016	0.014	-0.041
1.75	0.042				-0.178	-0.012	0.022	0.075	0.151
0.75	0.018				-0.115	-0.057	0.079	0.042	0.057
-0.25	-0.006						0.101	0.166	0.004
-1.25	-0.030				0.143	0.130	0.116	0.083	0.070
-2.25	-0.054				0.172	0.134	0.195	0.195	0.080
-2.75	-0.066		0.177	0.192					
-3.25	-0.077				0.115	0.170	0.036	0.036	0.120
-4.25	-0.101	0.042	0.070	0.023	0.128	0.216	0.131	0.131	0.177
-5.25	-0.125				0.163	0.055	0.085	0.085	0.025
-6.25	-0.149	0.131	0.085	0.131	0.214	0.154	0.098		0.010
-9.25	-0.220		0.131	0.076	0.040	0.085	0.148		
-12.25	-0.292		0.184	0.087	0.129	0.096			
-15.25	-0.363			0.131	0.083	0.131			

LINE COUNT = 905

RUN.SEQ
165.3

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.820	2.986	6.79	1505	968	538.1	455.4	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.440	0.083	0.523	-0.0206	-0.0131	-0.0338	0.2245	29.68	40.91	31.46	42.95	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.343	0.097	0.440	-0.0418	-0.0184	-0.0602	37.16	44.04	38.67	0.631	0.141
0.296	0.437	0.086	0.524	-0.0342	-0.0232	-0.0574	32.82	51.82	35.96	0.639	-0.002
0.500	0.506	0.093	0.599	-0.0282	-0.0277	-0.0559	30.57	54.94	34.34	0.598	-0.107
0.697	0.538	0.082	0.619	-0.0287	-0.0259	-0.0546	30.34	56.74	33.81	0.487	-0.137
0.894	0.550	0.022	0.572	-0.0462	-0.0155	-0.0616	33.39	96.53	35.78	0.328	-0.106

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.543	0.543	0.543	0.543	0.543
0.01			0.404		
0.02			0.329		
0.03			0.278		
0.04			0.244		
0.05	0.225	0.214	0.214	0.188	0.091
0.10	0.151	0.122	0.105	0.094	0.020
0.15	0.114	0.077	0.068	0.037	-0.024
0.20	0.096	0.043	0.041	0.023	-0.054
0.30	0.059	0.010	0.004	-0.008	-0.096
0.40	0.027	-0.015	-0.012	-0.022	-0.102
0.50	-0.002	-0.032	-0.010	-0.028	-0.087
0.55					
0.60	0.036	0.042	0.058	0.045	-0.007
0.65			0.110		
0.70	0.120	0.146	0.154	0.146	0.110
0.75	0.146	0.172	0.186	0.181	0.145
0.80	0.161	0.185	0.204	0.194	0.154
0.85	0.150	0.186	0.197	0.190	0.158
0.90	0.129	0.150	0.173	0.164	0.129
0.95	0.073	0.107	0.122	0.125	0.085
1.00	0.019	0.039	0.039	0.047	-0.017

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.449	0.357	0.273	0.219	0.237
0.003			-0.144		
0.006			-0.362	-0.392	
0.01	-0.275	-0.508	-0.619	-0.642	-0.554
0.02	-0.574	-0.870	-0.949	-0.933	-0.779
0.03	-0.657	-0.967	-1.113	-1.066	-0.859
0.04	-0.644	-1.107	-1.155	-1.058	-0.816
0.05	-0.642	-1.152	-1.239	-1.281	-1.010
0.06	-0.608	-1.110	-1.264	-1.229	-1.018
0.08	-0.587	-1.083	-1.234	-1.187	-0.986
0.10	-0.561	-0.881	-1.260	-1.200	-0.993
0.125	-0.526	-0.834	-1.247	-1.224	-1.001
0.15	-0.517	-0.766	-1.212	-1.241	-1.006
0.175	-0.514	-0.717	-0.859	-1.211	-1.016
0.20	-0.500	-0.665	-0.762	-1.198	-1.044
0.225	-0.489	-0.614	-0.712	-0.887	-1.021
0.25	-0.463	-0.574	-0.680	-0.720	-1.022
0.30	-0.445	-0.543	-0.631	-0.656	-0.909
0.35	-0.433	-0.526	-0.600	-0.643	-0.810
0.40	-0.439	-0.494	-0.576	-0.621	-0.722
0.45	-0.417	-0.456	-0.518	-0.589	-0.587
0.50	-0.387	-0.426	-0.479	-0.534	-0.463
0.55	-0.365	-0.385	-0.426	-0.461	-0.408
0.60	-0.333	-0.364	-0.370	-0.375	-0.371
0.65	-0.282	-0.332	-0.319	-0.318	-0.322
0.70	-0.255	-0.280	-0.283	-0.265	-0.277
0.75	-0.248	-0.238	-0.216	-0.213	-0.256
0.80	-0.195	-0.198	-0.184	-0.161	-0.219
0.85	-0.147	-0.137	-0.124	-0.114	-0.169
0.90	-0.086	-0.092	-0.055	-0.056	-0.130
0.95	-0.029	-0.006	-0.000	0.002	-0.082
1.00	0.019	0.039	0.039	0.047	-0.017

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	-0.006				-0.397				
11.35	0.270	0.083				-0.395				
12.35	0.294					-0.388		-0.362		
14.35	0.342	0.067				-0.398				
15.35	0.366	0.059				-0.412				-0.342
16.35	0.390	0.063				-0.404				0.286
17.35	0.413	0.049				-0.385		-0.362		-0.311
18.35	0.437	0.040				-0.392				-0.317
19.35	0.461	0.050				-0.381				-0.299
20.35	0.485	0.037				-0.380				-0.310
22.35	0.533	0.031				-0.350				-0.309
23.35	0.556	0.025				-0.363				-0.292
24.35	0.580	0.051				-0.324		-0.331		-0.303
25.35	0.604	0.054				-0.306				-0.289
26.35	0.628	0.066				-0.326				-0.290
27.35	0.652	0.074				-0.313				
30.35	0.723									-0.230
31.35	0.747		0.139	-0.234		-0.234	-0.213	-0.231	-0.221	
32.35	0.771		0.144	-0.219		-0.211	-0.209	-0.197		
33.35	0.795		0.135	-0.201		-0.196	-0.180	-0.173		
34.35	0.818		0.147	-0.181		-0.167	-0.168	-0.152		
35.35	0.842		0.142	-0.150		-0.175	-0.156	-0.167		
36.35	0.866		0.133	-0.139		-0.147	-0.139	-0.144		
37.35	0.890		0.122	-0.104		-0.112	-0.134	-0.127		
38.35	0.914		0.103	-0.073		-0.101	-0.105	-0.107		
39.35	0.938		0.100	-0.034	-0.049	-0.075	-0.088	-0.094		
40.35	0.961		0.070	-0.001	-0.042	-0.048	-0.075	-0.096		
41.35	0.985		0.054	0.025	-0.002	-0.021	-0.059	-0.069		
42.35	1.009		0.055	0.042	0.022	-0.015	-0.040	-0.149		
44.85	1.069		0.027	0.005	0.021	-0.006	-0.042	-0.066		
45.85	1.092		0.038	0.018	0.012					
46.85	1.116		0.038	0.023	0.012					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.295	-0.206	-0.149			
13.75	0.328		-0.276	-0.263	-0.214	-0.147			
10.75	0.256		-0.311	-0.303	-0.221	-0.144	-0.069	-0.149	
7.75	0.185	-0.362	-0.362	-0.331	-0.213	-0.137	-0.058		-0.035
6.75	0.161				-0.231	-0.139	-0.059	-0.040	-0.042
5.75	0.137	-0.387	-0.390	-0.340	-0.236	-0.155	-0.035	-0.044	-0.013
4.75	0.113				-0.235	-0.155	-0.046	-0.038	-0.021
4.25	0.101	-0.388	-0.385	-0.324					
3.75	0.089					-0.234	-0.147	-0.021	-0.015
2.75	0.066					-0.235	-0.133	-0.029	-0.008
1.75	0.042					-0.234	-0.026	-0.013	0.010
0.75	0.018					-0.234	-0.139	-0.002	0.022
-0.25	-0.006							0.025	0.042
-1.25	-0.030						0.139	0.133	0.054
-2.25	-0.054						0.140	0.134	0.066
-2.75	-0.066			0.049	0.051				
-3.25	-0.077						0.107	0.117	0.073
-4.25	-0.101	0.085	0.045	0.045	0.099	0.113	0.082	0.066	0.048
-5.25	-0.125						0.109	0.107	0.080
-6.25	-0.149	0.089	0.056	0.062	0.087	0.119	0.079		-0.075
-9.25	-0.220		0.065	0.052	0.075	0.081	0.083		
-12.25	-0.292		0.060	0.054	0.080	0.074			
-15.25	-0.363			0.057	0.077	0.069			

RUN.SEQ
166.3

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.893	4.347	9.89	2163	1288	548.2	719.2	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.484	0.079	0.563	-0.0505	-0.0110	-0.0615	0.2402	35.44	38.96	35.93	42.70	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.377	0.108	0.485	-0.0684	-0.0223	-0.0907	43.15	45.61	43.70	0.695	0.105
0.296	0.463	0.093	0.556	-0.0572	-0.0249	-0.0821	37.36	51.73	39.77	0.678	-0.034
0.500	0.550	0.081	0.631	-0.0579	-0.0271	-0.0850	35.54	58.56	38.49	0.630	-0.138
0.697	0.631	0.065	0.695	-0.0754	-0.0224	-0.0978	36.96	59.63	39.07	0.547	-0.177
0.894	0.595	-0.021	0.574	-0.1051	-0.0040	-0.1091	42.65	6.085	44.01	0.330	-0.122

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.552	0.552	0.552	0.552	0.552
0.01			0.367		
0.02			0.301		
0.03			0.257		
0.04			0.222		
0.05	0.244	0.206	0.180	0.163	0.029
0.10	0.171	0.125	0.087	0.073	-0.030
0.15	0.129	0.089	0.053	0.024	-0.072
0.20	0.101	0.061	0.031	0.001	-0.096
0.30	0.062	0.017	-0.010	-0.033	-0.152
0.40	0.029	-0.003	-0.033	-0.044	-0.146
0.50	0.002	-0.025	-0.026	-0.049	-0.121
0.55					
0.60	0.038	0.033	0.044	0.034	-0.029
0.65			0.100		
0.70	0.129	0.157	0.151	0.139	0.096
0.75	0.165	0.190	0.188	0.174	0.127
0.80	0.191	0.196	0.201	0.189	0.126
0.85	0.186	0.197	0.198	0.184	0.127
0.90	0.148	0.159	0.170	0.159	0.096
0.95	0.094	0.103	0.119	0.099	0.004
1.00	0.023	0.055	0.055	-0.006	-0.246

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.528	0.451	0.382	0.347	0.362
0.003			0.033		
0.006			-0.102	-0.186	
0.01	-0.132	-0.313	-0.385	-0.402	-0.301
0.02	-0.425	-0.643	-0.692	-0.662	-0.498
0.03	-0.544	-0.807	-0.843	-0.792	-0.563
0.04	-0.523	-0.881	-0.907	-0.780	-0.546
0.05	-0.557	-0.908	-0.939	-0.945	-0.776
0.06	-0.531	-0.905	-0.984	-0.945	-0.773
0.08	-0.504	-0.915	-0.989	-0.915	-0.761
0.10	-0.487	-0.912	-1.032	-0.957	-0.764
0.125	-0.470	-0.729	-1.029	-0.983	-0.782
0.15	-0.454	-0.701	-1.035	-1.012	-0.806
0.175	-0.452	-0.672	-1.018	-1.009	-0.824
0.20	-0.451	-0.628	-0.971	-1.014	-0.850
0.225	-0.437	-0.591	-0.800	-1.017	-0.878
0.25	-0.426	-0.565	-0.713	-1.002	-0.889
0.30	-0.436	-0.532	-0.653	-1.019	-0.961
0.35	-0.427	-0.515	-0.615	-0.991	-0.943
0.40	-0.432	-0.513	-0.623	-0.758	-0.570
0.45	-0.441	-0.507	-0.613	-0.729	-0.547
0.50	-0.439	-0.502	-0.607	-0.728	-0.550
0.55	-0.441	-0.500	-0.604	-0.719	-0.539
0.60	-0.444	-0.496	-0.593	-0.635	-0.533
0.65	-0.423	-0.482	-0.588	-0.521	-0.537
0.70	-0.468	-0.491	-0.577	-0.387	-0.514
0.75	-0.447	-0.477	-0.431	-0.355	-0.491
0.80	-0.363	-0.291	-0.169	-0.286	-0.432
0.85	-0.225	-0.116	-0.089	-0.220	-0.381
0.90	-0.109	-0.050	-0.033	-0.140	-0.335
0.95	-0.022	0.014	0.019	-0.065	-0.309
1.00	0.023	0.055	0.055	-0.006	-0.246

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.062				-0.343				
11.35	0.270	0.093				-0.353				
12.35	0.294					-0.373		-0.306		
14.35	0.342	0.081				-0.380				
15.35	0.366	0.068				-0.380				-0.314
16.35	0.390	0.059				-0.386				0.734
17.35	0.413	0.060				-0.385		-0.335		-0.300
18.35	0.437	0.044				-0.391				-0.315
19.35	0.461	0.036				-0.386				-0.295
20.35	0.485	0.030				-0.397				-0.303
22.35	0.533	0.028				-0.390				-0.324
23.35	0.556	0.034				-0.410				-0.330
24.35	0.580	0.048				-0.408		-0.380		-0.340
25.35	0.604	0.052				-0.401				-0.346
26.35	0.628	0.064				-0.401				-0.354
27.35	0.652	0.071				-0.399				
30.35	0.723									-0.347
31.35	0.747			0.140	-0.389		-0.378	-0.376	-0.378	-0.330
32.35	0.771			0.153	-0.324		-0.372		-0.362	-0.321
33.35	0.795			0.156	-0.346		-0.353		-0.337	-0.281
34.35	0.818			0.158	-0.254		-0.349		-0.247	-0.240
35.35	0.842			0.168	-0.187		-0.223		-0.180	-0.191
36.35	0.866			0.148	-0.145		-0.192		-0.159	-0.149
37.35	0.890			0.130	-0.090		-0.137		-0.176	-0.119
38.35	0.914			0.101	-0.063		-0.126		-0.152	-0.106
39.35	0.938			0.101	-0.038	-0.031	-0.074		-0.104	-0.104
40.35	0.961			0.085	-0.007	-0.002	-0.053		-0.049	-0.087
41.35	0.985			0.072	0.039	0.028	-0.017		-0.053	-0.076
42.35	1.009			0.076	0.066	0.040	0.006		-0.042	-0.105
44.85	1.069			0.060	0.078	0.030	0.007		-0.006	-0.037
45.85	1.092			0.046	0.070	0.020				
46.85	1.116			0.034	0.018	0.034				

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.302	-0.291	-0.221			
13.75	0.328			-0.255	-0.309	-0.321	-0.196		
10.75	0.256			-0.300	-0.340	-0.330	-0.149	-0.076	-0.105
7.75	0.185	-0.306	-0.335	-0.380	-0.376	-0.167	-0.052		-0.026
6.75	0.161				-0.378	-0.159	-0.053	-0.042	-0.006
5.75	0.137	-0.335	-0.365	-0.404	-0.397	-0.163	-0.014	0.001	-0.004
4.75	0.113				-0.415	-0.183	-0.037	0.002	0.009
4.25	0.101	-0.373	-0.385	-0.408					
3.75	0.089				-0.378	-0.192	-0.017	0.006	0.007
2.75	0.066				-0.359	-0.174	-0.021	0.020	0.037
1.75	0.042				-0.395	-0.169	0.007	0.016	0.049
0.75	0.018				-0.389	-0.145	0.028	0.040	0.030
-0.25	-0.006						0.039	0.066	0.078
-1.25	-0.030				0.140	0.148	0.072	0.076	0.060
-2.25	-0.054				0.121	0.140	0.087	0.076	0.052
-2.75	-0.066			0.060	0.048				
-3.25	-0.077				0.126	0.128	0.090	0.087	0.050
-4.25	-0.101	0.089	0.054	0.047	0.108	0.134	0.090	0.090	0.065
-5.25	-0.125				0.112	0.127	0.092	0.086	0.066
-6.25	-0.149	0.097	0.067	0.051	0.123	0.118	0.085		0.001
-9.25	-0.220		0.076	0.075	0.106	0.084	0.090		
-12.25	-0.292		0.076	0.066	0.078	0.076			
-15.25	-0.363			0.060	0.075	0.085			

RUN.SEQ
167.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.877	4.386	9.98	2211	1340	550.6	721.2	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.464	0.089	0.553	-0.0348	-0.0148	-0.0496	0.2364	32.50	41.66	33.97	42.76	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.369	0.104	0.473	-0.0592	-0.0214	-0.0806	41.04	45.51	42.03	0.679	0.119
0.296	0.459	0.097	0.556	-0.0490	-0.0260	-0.0750	35.68	51.66	38.48	0.679	-0.023
0.500	0.515	0.101	0.616	-0.0371	-0.0316	-0.0687	32.20	56.24	36.14	0.616	-0.121
0.697	0.576	0.088	0.665	-0.0427	-0.0293	-0.0720	32.42	58.10	35.83	0.523	-0.156
0.894	0.587	0.005	0.593	-0.0852	-0.0112	-0.0964	39.52	231.8	41.27	0.340	-0.121

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.550	0.550	0.550	0.550	0.550
0.01			0.389		
0.02			0.317		
0.03			0.269		
0.04			0.237		
0.05	0.243	0.208	0.204	0.190	0.063
0.10	0.165	0.130	0.113	0.097	0.009
0.15	0.124	0.084	0.074	0.039	-0.027
0.20	0.087	0.060	0.050	0.027	-0.056
0.30	0.057	0.028	0.011	-0.008	-0.128
0.40	0.028	0.005	-0.002	-0.024	-0.128
0.50	0.004	-0.018	-0.008	-0.022	-0.106
0.55					
0.60	0.042	0.044	0.064	0.052	-0.019
0.65			0.118		
0.70	0.133	0.151	0.169	0.150	0.101
0.75	0.156	0.190	0.204	0.186	0.140
0.80	0.172	0.203	0.215	0.200	0.156
0.85	0.170	0.199	0.216	0.200	0.152
0.90	0.146	0.154	0.185	0.197	0.123
0.95	0.091	0.114	0.143	0.148	0.062
1.00	0.030	0.052	0.067	0.073	-0.118

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.507	0.433	0.355	0.320	0.336
0.003			-0.006		
0.006			-0.203	-0.233	
0.01	-0.173	-0.355	-0.433	-0.463	-0.373
0.02	-0.465	-0.692	-0.751	-0.729	-0.581
0.03	-0.572	-0.827	-0.905	-0.859	-0.658
0.04	-0.564	-0.934	-0.965	-0.851	-0.601
0.05	-0.588	-0.956	-1.006	-1.012	-0.831
0.06	-0.554	-0.950	-1.048	-1.012	-0.834
0.08	-0.527	-0.965	-1.046	-0.987	-0.818
0.10	-0.509	-0.916	-1.088	-1.012	-0.822
0.125	-0.486	-0.756	-1.081	-1.036	-0.846
0.15	-0.475	-0.740	-1.083	-1.061	-0.862
0.175	-0.474	-0.701	-1.060	-1.060	-0.880
0.20	-0.464	-0.650	-0.938	-1.065	-0.904
0.225	-0.454	-0.615	-0.754	-1.063	-0.929
0.25	-0.441	-0.581	-0.696	-1.050	-0.934
0.30	-0.451	-0.543	-0.642	-1.048	-1.009
0.35	-0.435	-0.532	-0.613	-0.769	-0.952
0.40	-0.444	-0.525	-0.621	-0.725	-0.590
0.45	-0.440	-0.520	-0.612	-0.731	-0.586
0.50	-0.442	-0.514	-0.600	-0.727	-0.583
0.55	-0.450	-0.513	-0.588	-0.714	-0.560
0.60	-0.444	-0.499	-0.570	-0.512	-0.530
0.65	-0.423	-0.478	-0.471	-0.254	-0.501
0.70	-0.449	-0.442	-0.244	-0.216	-0.456
0.75	-0.351	-0.317	-0.163	-0.203	-0.401
0.80	-0.253	-0.176	-0.126	-0.171	-0.350
0.85	-0.164	-0.115	-0.082	-0.134	-0.296
0.90	-0.088	-0.059	-0.027	-0.028	-0.234
0.95	-0.017	0.006	0.025	0.033	-0.179
1.00	0.030	0.052	0.067	0.073	-0.118

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.079				-0.352				
11.35	0.270	0.101				-0.360				
12.35	0.294					-0.369		-0.330		
14.35	0.342	0.092				-0.374				
15.35	0.366	0.074				-0.370				-0.313
16.35	0.390	0.061				-0.373				0.560
17.35	0.413	0.073				-0.385		-0.353		-0.303
18.35	0.437	0.060				-0.400				-0.314
19.35	0.461	0.050				-0.386				-0.315
20.35	0.485	0.037				-0.383				-0.315
22.35	0.533	0.041				-0.382				-0.329
23.35	0.556	0.059				-0.393				-0.344
24.35	0.580	0.059				-0.391		-0.375		-0.345
25.35	0.604	0.064				-0.386				-0.348
26.35	0.628	0.067				-0.385				-0.349
27.35	0.652	0.089				-0.378				
30.35	0.723									-0.323
31.35	0.747		0.150		-0.354		-0.281	-0.318	-0.283	-0.340
32.35	0.771		0.155		-0.303		-0.250		-0.278	-0.301
33.35	0.795		0.193		-0.246		-0.224		-0.251	-0.257
34.35	0.818		0.181		-0.222		-0.193		-0.186	-0.216
35.35	0.842		0.157		-0.174		-0.159		-0.168	-0.171
36.35	0.866		0.148		-0.131		-0.145		-0.153	-0.161
37.35	0.890		0.131		-0.087		-0.121		-0.096	-0.142
38.35	0.914		0.113		-0.062		-0.077		-0.073	-0.134
39.35	0.938		0.116	-0.016	-0.028		-0.051		-0.077	-0.097
40.35	0.961		0.105	-0.006	-0.012		-0.031		-0.056	-0.077
41.35	0.985		0.085	0.041	0.016		-0.020		-0.031	-0.084
42.35	1.009		0.084	0.059	0.036		-0.005		-0.012	-0.120
44.85	1.069		0.058	0.057	0.048		0.007		-0.005	-0.069
45.85	1.092		0.060	0.049	0.050					
46.85	1.116		0.046	0.030	0.035					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.286	-0.271	-0.177			
13.75	0.328		-0.273	-0.315	-0.291	-0.144			
10.75	0.256		-0.303	-0.345	-0.340	-0.161	-0.084	-0.120	
7.75	0.185	-0.330	-0.353	-0.375	-0.318	-0.119	-0.056		-0.025
6.75	0.161				-0.283	-0.153	-0.031	-0.012	-0.005
5.75	0.137	-0.365	-0.381	-0.395	-0.322	-0.140	-0.036	-0.012	-0.027
4.75	0.113				-0.317	-0.129	-0.030	-0.006	-0.012
4.25	0.101	-0.369	-0.385	-0.391					
3.75	0.089				-0.281	-0.145	-0.020	-0.005	0.007
2.75	0.066				-0.346	-0.127	0.008	0.005	0.016
1.75	0.042				-0.293	-0.166	0.016	0.017	0.036
0.75	0.018				-0.354	-0.131	0.016	0.036	0.048
-0.25	-0.006						0.041	0.059	0.057
-1.25	-0.030					0.150	0.148	0.085	0.084	0.058
-2.25	-0.054					0.155	0.141	0.099	0.098	0.077
-2.75	-0.066			0.073	0.059					
-3.25	-0.077					0.130	0.150	0.108	0.106	0.068
-4.25	-0.101	0.095	0.062	0.053	0.119	0.129	0.110	0.108	0.068
-5.25	-0.125				0.113	0.103	0.095	0.101	0.064
-6.25	-0.149	0.098	0.068	0.067	0.109	0.102	0.096		0.014
-9.25	-0.220		0.079	0.090	0.091	0.095	0.095		
-12.25	-0.292		0.082	0.086	0.075	0.083			
-15.25	-0.363			0.081	0.080	0.076			

RUN.SEQ
168.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.860	4.364	9.93	2227	1374	552.5	711.5	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.464	0.083	0.547	-0.0305	-0.0129	-0.0434	0.2345	31.58	40.59	32.95	42.88	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.369	0.099	0.468	-0.0565	-0.0186	-0.0752	40.33	43.75	41.06	0.671	0.127
0.296	0.453	0.095	0.548	-0.0429	-0.0252	-0.0681	34.48	51.54	37.44	0.668	-0.014
0.500	0.512	0.089	0.602	-0.0316	-0.0290	-0.0606	31.18	57.40	35.07	0.602	-0.112
0.697	0.588	0.077	0.666	-0.0398	-0.0260	-0.0658	31.76	58.70	34.88	0.524	-0.152
0.894	0.588	0.007	0.594	-0.0653	-0.0138	-0.0790	36.11	226.9	38.30	0.341	-0.115

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.548	0.548	0.548	0.548	0.548
0.01			0.395		
0.02			0.328		
0.03			0.277		
0.04			0.233		
0.05	0.237	0.211	0.189	0.182	0.061
0.10	0.164	0.128	0.099	0.085	0.002
0.15	0.127	0.086	0.058	0.034	-0.045
0.20	0.095	0.058	0.037	0.012	-0.072
0.30	0.061	0.024	-0.006	-0.017	-0.127
0.40	0.030	-0.002	-0.019	-0.028	-0.124
0.50	-0.004	-0.023	-0.016	-0.040	-0.105
0.55					
0.60	0.029	0.046	0.058	0.030	-0.017
0.65			0.116		
0.70	0.118	0.149	0.154	0.148	0.098
0.75	0.152	0.184	0.188	0.187	0.136
0.80	0.164	0.197	0.201	0.203	0.161
0.85	0.162	0.197	0.199	0.199	0.166
0.90	0.134	0.166	0.181	0.174	0.142
0.95	0.080	0.106	0.136	0.122	0.074
1.00	0.007	0.040	0.054	0.050	-0.070

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.490	0.411	0.333	0.290	0.305
0.003			-0.043		
0.006			-0.247	-0.276	
0.01	-0.199	-0.398	-0.485	-0.512	-0.424
0.02	-0.495	-0.741	-0.805	-0.781	-0.632
0.03	-0.598	-0.878	-0.962	-0.915	-0.712
0.04	-0.586	-0.986	-1.024	-0.910	-0.668
0.05	-0.602	-1.010	-1.064	-1.063	-0.870
0.06	-0.574	-0.997	-1.105	-1.073	-0.886
0.08	-0.544	-1.009	-1.098	-1.048	-0.868
0.10	-0.528	-0.897	-1.135	-1.057	-0.874
0.125	-0.496	-0.779	-1.130	-1.089	-0.897
0.15	-0.486	-0.756	-1.128	-1.114	-0.909
0.175	-0.488	-0.707	-1.101	-1.111	-0.924
0.20	-0.483	-0.659	-0.886	-1.115	-0.946
0.225	-0.467	-0.621	-0.752	-1.112	-0.972
0.25	-0.453	-0.597	-0.698	-1.094	-0.978
0.30	-0.458	-0.563	-0.665	-1.062	-1.054
0.35	-0.447	-0.534	-0.628	-0.756	-1.123
0.40	-0.446	-0.528	-0.630	-0.743	-0.779
0.45	-0.449	-0.508	-0.614	-0.763	-0.662
0.50	-0.446	-0.509	-0.594	-0.755	-0.628
0.55	-0.437	-0.491	-0.576	-0.684	-0.582
0.60	-0.423	-0.471	-0.434	-0.459	-0.523
0.65	-0.403	-0.430	-0.314	-0.259	-0.427
0.70	-0.382	-0.330	-0.249	-0.215	-0.349
0.75	-0.293	-0.237	-0.170	-0.185	-0.284
0.80	-0.238	-0.186	-0.138	-0.152	-0.210
0.85	-0.176	-0.118	-0.084	-0.107	-0.149
0.90	-0.105	-0.059	-0.040	-0.037	-0.119
0.95	-0.035	-0.010	0.018	0.023	-0.107
1.00	0.007	0.040	0.054	0.050	-0.070

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.076				-0.369				
11.35	0.270	0.094				-0.378				
12.35	0.294					-0.387		-0.336		
14.35	0.342	0.080				-0.392				
15.35	0.366	0.063				-0.394				-0.326
16.35	0.390	0.058				-0.384				0.443
17.35	0.413	0.047				-0.396		-0.359		-0.327
18.35	0.437	0.034				-0.401				-0.337
19.35	0.461	0.034				-0.407				-0.338
20.35	0.485	0.051				-0.404				-0.339
22.35	0.533	0.037				-0.386				-0.351
23.35	0.556	0.041				-0.391				-0.323
24.35	0.580	0.044				-0.381		-0.379		-0.340
25.35	0.604	0.053				-0.366				-0.357
26.35	0.628	0.060				-0.335				-0.346
27.35	0.652	0.075				-0.349				
30.35	0.723									-0.277
31.35	0.747		0.146		-0.255		-0.275	-0.221	-0.222	-0.275
32.35	0.771		0.160		-0.218		-0.201		-0.214	-0.255
33.35	0.795		0.154		-0.199		-0.186		-0.200	-0.227
34.35	0.818		0.150		-0.188		-0.186		-0.187	-0.209
35.35	0.842		0.147		-0.165		-0.148		-0.162	-0.170
36.35	0.866		0.146		-0.124		-0.114		-0.144	-0.147
37.35	0.890		0.144		-0.096		-0.098		-0.141	-0.144
38.35	0.914		0.127		-0.052		-0.108		-0.088	-0.128
39.35	0.938		0.120	-0.037	-0.014		-0.075		-0.092	-0.104
40.35	0.961		0.099	-0.007	-0.004		-0.053		-0.054	-0.091
41.35	0.985		0.085	0.034	0.019		-0.012		-0.033	-0.086
42.35	1.009		0.072	0.049	0.027		0.008		-0.017	-0.114
44.85	1.069		0.054	0.036	0.029		0.018		-0.010	-0.063
45.85	1.092		0.046	0.034	0.026					
46.85	1.116		0.035	0.034	0.025					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.281	-0.232	-0.135			
13.75	0.328		-0.277	-0.309	-0.237	-0.158			
10.75	0.256		-0.327	-0.340	-0.275	-0.147	-0.086	-0.114	
7.75	0.185	-0.336	-0.359	-0.379	-0.221	-0.171	-0.054		-0.015
6.75	0.161				-0.222	-0.144	-0.033	-0.017	-0.010
5.75	0.137	-0.369	-0.388	-0.402	-0.260	-0.145	-0.025	-0.019	-0.020
4.75	0.113				-0.263	-0.145	-0.041	-0.020	-0.002
4.25	0.101	-0.387	-0.396	-0.381					
3.75	0.089				-0.275	-0.114	-0.012	0.008	0.018
2.75	0.066				-0.243	-0.152	-0.023	-0.002	0.016
1.75	0.042				-0.261	-0.157	-0.003	0.017	0.036
0.75	0.018				-0.255	-0.124	0.019	0.027	0.029
-0.25	-0.006						0.034	0.049	0.036
-1.25	-0.030					0.146	0.146	0.085	0.072	0.054
-2.25	-0.054					0.138	0.147	0.110	0.076	0.061
-2.75	-0.066			0.047	0.044					
-3.25	-0.077					0.121	0.127	0.114	0.073	0.057
-4.25	-0.101	0.086	0.061	0.041	0.123	0.113	0.094	0.085	0.071
-5.25	-0.125				0.124	0.108	0.092	0.099	0.072
-6.25	-0.149	0.097	0.072	0.065	0.105	0.107	0.085		0.017
-9.25	-0.220		0.093	0.075	0.090	0.085	0.085		
-12.25	-0.292		0.068	0.068	0.092	0.075			
-15.25	-0.363			0.063	0.082	0.069			

RUN.SEQ
169.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.850	4.377	9.96	2253	1404	553.9	710.4	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.451	0.088	0.540	-0.0253	-0.0140	-0.0393	0.2319	30.61	40.82	32.28	42.98	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.356	0.108	0.464	-0.0491	-0.0216	-0.0707	38.79	44.99	40.24	0.665	0.134
0.296	0.432	0.096	0.528	-0.0342	-0.0254	-0.0596	32.91	51.48	36.28	0.644	-0.004
0.500	0.507	0.097	0.603	-0.0301	-0.0300	-0.0601	30.93	56.03	34.96	0.603	-0.111
0.697	0.568	0.081	0.649	-0.0318	-0.0275	-0.0593	30.60	59.19	34.14	0.511	-0.145
0.894	0.593	0.016	0.609	-0.0601	-0.0164	-0.0765	35.13	127.3	37.55	0.350	-0.117

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.547	0.547	0.547	0.547	0.547
0.01			0.383		
0.02			0.312		
0.03			0.268		
0.04			0.249		
0.05	0.244	0.223	0.201	0.179	0.058
0.10	0.173	0.125	0.105	0.087	0.007
0.15	0.135	0.079	0.071	0.033	-0.033
0.20	0.096	0.056	0.050	0.012	-0.062
0.30	0.055	0.025	0.006	-0.021	-0.115
0.40	0.036	0.000	-0.009	-0.029	-0.112
0.50	0.017	-0.016	-0.009	-0.027	-0.087
0.55					
0.60	0.048	0.037	0.076	0.046	-0.011
0.65			0.128		
0.70	0.128	0.147	0.162	0.146	0.112
0.75	0.170	0.183	0.197	0.189	0.148
0.80	0.178	0.199	0.208	0.207	0.164
0.85	0.169	0.204	0.204	0.203	0.167
0.90	0.135	0.170	0.171	0.174	0.154
0.95	0.087	0.100	0.129	0.129	0.089
1.00	0.019	0.056	0.052	0.052	-0.071

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.489	0.403	0.320	0.276	0.232
0.003			-0.063		
0.006			-0.271	-0.305	
0.01	-0.211	-0.422	-0.513	-0.547	-0.452
0.02	-0.507	-0.770	-0.836	-0.818	-0.667
0.03	-0.610	-0.907	-0.997	-0.953	-0.751
0.04	-0.597	-1.014	-1.061	-0.950	-0.710
0.05	-0.616	-1.034	-1.105	-1.087	-0.895
0.06	-0.583	-1.012	-1.135	-1.107	-0.913
0.08	-0.550	-1.037	-1.128	-1.080	-0.899
0.10	-0.529	-0.857	-1.163	-1.089	-0.905
0.125	-0.511	-0.785	-1.160	-1.119	-0.928
0.15	-0.502	-0.769	-1.154	-1.146	-0.936
0.175	-0.492	-0.709	-1.090	-1.139	-0.953
0.20	-0.487	-0.667	-0.823	-1.145	-0.978
0.225	-0.471	-0.613	-0.731	-1.139	-0.988
0.25	-0.455	-0.594	-0.688	-1.117	-0.996
0.30	-0.459	-0.570	-0.641	-0.916	-1.066
0.35	-0.453	-0.515	-0.610	-0.748	-1.138
0.40	-0.448	-0.509	-0.617	-0.755	-0.953
0.45	-0.434	-0.500	-0.596	-0.742	-0.672
0.50	-0.431	-0.466	-0.561	-0.688	-0.642
0.55	-0.428	-0.429	-0.488	-0.492	-0.549
0.60	-0.404	-0.406	-0.393	-0.325	-0.458
0.65	-0.354	-0.334	-0.303	-0.252	-0.430
0.70	-0.298	-0.266	-0.255	-0.210	-0.321
0.75	-0.264	-0.220	-0.199	-0.188	-0.224
0.80	-0.218	-0.172	-0.163	-0.147	-0.170
0.85	-0.150	-0.113	-0.109	-0.096	-0.137
0.90	-0.091	-0.057	-0.041	-0.039	-0.108
0.95	-0.014	0.010	0.021	0.018	-0.097
1.00	0.019	0.056	0.052	0.052	-0.071

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.070				-0.368				
11.35	0.270	0.085				-0.383				
12.35	0.294					-0.396		-0.344		
14.35	0.342	0.091				-0.397				
15.35	0.366	0.070				-0.393				-0.334
16.35	0.390	0.052				-0.386				0.341
17.35	0.413	0.063				-0.391		-0.374		-0.320
18.35	0.437	0.041				-0.399				-0.332
19.35	0.461	0.038				-0.401				-0.335
20.35	0.485	0.033				-0.389				-0.331
22.35	0.533	0.036				-0.363				-0.341
23.35	0.556	0.045				-0.387				-0.320
24.35	0.580	0.042				-0.391		-0.341		-0.329
25.35	0.604	0.049				-0.368				-0.333
26.35	0.628	0.062				-0.335				-0.286
27.35	0.652	0.082				-0.331				
30.35	0.723									-0.225
31.35	0.747		0.150		-0.268		-0.262	-0.243	-0.259	-0.201
32.35	0.771		0.158		-0.232		-0.218		-0.235	-0.194
33.35	0.795		0.163		-0.212		-0.210		-0.220	-0.198
34.35	0.818		0.154		-0.184		-0.184		-0.201	-0.164
35.35	0.842		0.145		-0.143		-0.156		-0.171	-0.160
36.35	0.866		0.145		-0.127		-0.143		-0.152	-0.147
37.35	0.890		0.140		-0.099		-0.115		-0.123	-0.115
38.35	0.914		0.111		-0.063		-0.090		-0.105	-0.095
39.35	0.938		0.101	-0.024	-0.043		-0.070		-0.089	-0.073
40.35	0.961		0.076	-0.009	-0.011		-0.051		-0.052	-0.087
41.35	0.985		0.061	0.015	0.021		-0.035		-0.047	-0.080
42.35	1.009		0.060	0.062	0.038		-0.017		-0.034	-0.111
44.85	1.069		0.025	0.057	0.027		0.006		-0.019	-0.072
45.85	1.092		0.022	0.039	0.015					
46.85	1.116		0.041	0.027	0.046					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.299	-0.248	-0.163			
13.75	0.328			-0.295	-0.301	-0.252	-0.147		
10.75	0.256			-0.320	-0.329	-0.201	-0.147	-0.080	-0.111
7.75	0.185	-0.344	-0.374	-0.341	-0.243	-0.135	-0.060		-0.028
6.75	0.161					-0.259	-0.152	-0.047	-0.034 -0.019
5.75	0.137	-0.386	-0.385	-0.356	-0.265	-0.129	-0.048	-0.032	-0.017
4.75	0.113					-0.276	-0.135	-0.044	-0.021 -0.019
4.25	0.101	-0.396	-0.391	-0.391					
3.75	0.089					-0.262	-0.143	-0.035	-0.017 0.006
2.75	0.066					-0.270	-0.136	-0.009	-0.004 0.007
1.75	0.042					-0.259	-0.177	0.002	0.007 0.014
0.75	0.018					-0.268	-0.127	0.021	0.038 0.027
-0.25	-0.006							0.015	0.062 0.057
-1.25	-0.030							0.150	0.145 0.061 0.060 0.025
-2.25	-0.054							0.116	0.125 0.100 0.094 0.049
-2.75	-0.066			0.063	0.042				
-3.25	-0.077							0.102	0.114 0.077 0.078 0.045
-4.25	-0.101	0.090	0.052	0.051				0.095	0.121 0.082 0.077 0.061
-5.25	-0.125							0.102	0.112 0.075 0.087 0.058
-6.25	-0.149	0.084	0.057	0.045				0.086	0.105 0.078 0.013
-9.25	-0.220		0.066	0.065				0.078	0.083 0.074
-12.25	-0.292		0.087	0.067				0.076	0.081
-15.25	-0.363			0.065				0.062	0.074

RUN.SEQ
170.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.837	4.382	9.97	2276	1438	554.8	705.1	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.453	0.084	0.538	-0.0251	-0.0127	-0.0378	0.2313	30.54	40.06	32.04	43.01	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.355	0.107	0.462	-0.0469	-0.0223	-0.0692	38.22	45.73	39.97	0.663	0.136
0.296	0.429	0.098	0.527	-0.0330	-0.0260	-0.0590	32.68	51.59	36.19	0.643	-0.004
0.500	0.512	0.086	0.597	-0.0304	-0.0270	-0.0574	30.95	56.39	34.60	0.597	-0.108
0.697	0.578	0.069	0.647	-0.0346	-0.0229	-0.0575	30.99	58.18	33.89	0.509	-0.144
0.894	0.599	0.014	0.613	-0.0570	-0.0159	-0.0729	34.52	138.9	36.90	0.352	-0.116

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.545	0.545	0.545	0.545	0.545
0.01			0.396		
0.02			0.317		
0.03			0.267		
0.04			0.243		
0.05	0.241	0.217	0.197	0.188	0.087
0.10	0.163	0.131	0.098	0.090	-0.003
0.15	0.124	0.081	0.059	0.029	-0.043
0.20	0.096	0.056	0.035	0.006	-0.067
0.30	0.059	0.016	-0.002	-0.026	-0.127
0.40	0.021	0.003	-0.021	-0.038	-0.119
0.50	0.0	-0.008	-0.021	-0.052	-0.093
0.55					
0.60	0.051	0.056	0.049	0.017	-0.014
0.65			0.112		
0.70	0.131	0.155	0.161	0.134	0.111
0.75	0.165	0.186	0.186	0.180	0.151
0.80	0.171	0.196	0.195	0.200	0.163
0.85	0.182	0.191	0.198	0.193	0.158
0.90	0.137	0.152	0.163	0.156	0.135
0.95	0.091	0.117	0.113	0.112	0.097
1.00	0.026	0.044	0.054	0.028	-0.024

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.458	0.393	0.309	0.262	0.272
0.003			-0.081		
0.006			-0.294	-0.332	
0.01	-0.228	-0.443	-0.541	-0.575	-0.489
0.02	-0.523	-0.798	-0.868	-0.848	-0.703
0.03	-0.623	-0.938	-1.027	-0.984	-0.791
0.04	-0.616	-1.044	-1.093	-0.986	-0.745
0.05	-0.631	-1.055	-1.140	-1.107	-0.921
0.06	-0.591	-1.032	-1.168	-1.144	-0.942
0.08	-0.566	-1.059	-1.159	-1.114	-0.926
0.10	-0.546	-0.877	-1.192	-1.121	-0.932
0.125	-0.527	-0.800	-1.188	-1.154	-0.948
0.15	-0.515	-0.768	-1.182	-1.180	-0.958
0.175	-0.510	-0.701	-1.062	-1.173	-0.979
0.20	-0.497	-0.656	-0.789	-1.172	-1.001
0.225	-0.486	-0.601	-0.718	-1.164	-1.014
0.25	-0.467	-0.573	-0.684	-1.141	-1.023
0.30	-0.467	-0.543	-0.665	-0.828	-1.089
0.35	-0.450	-0.517	-0.627	-0.729	-1.181
0.40	-0.439	-0.499	-0.616	-0.735	-0.963
0.45	-0.433	-0.474	-0.581	-0.743	-0.683
0.50	-0.426	-0.462	-0.510	-0.657	-0.638
0.55	-0.410	-0.438	-0.485	-0.492	-0.560
0.60	-0.375	-0.381	-0.416	-0.360	-0.452
0.65	-0.331	-0.313	-0.314	-0.283	-0.353
0.70	-0.300	-0.253	-0.260	-0.239	-0.254
0.75	-0.246	-0.223	-0.206	-0.198	-0.214
0.80	-0.205	-0.174	-0.168	-0.158	-0.190
0.85	-0.154	-0.112	-0.115	-0.107	-0.160
0.90	-0.094	-0.060	-0.052	-0.059	-0.128
0.95	-0.023	0.003	0.016	-0.005	-0.083
1.00	0.026	0.044	0.054	0.028	-0.024

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR								
10.35	0.247	0.069			-0.401				
11.35	0.270	0.095			-0.406				
12.35	0.294				-0.409		-0.346		
14.35	0.342	0.077			-0.412				
15.35	0.366	0.077			-0.411				-0.335
16.35	0.390	0.068			-0.407				0.247
17.35	0.413	0.041			-0.405		-0.365		-0.324
18.35	0.437	0.046			-0.408				-0.330
19.35	0.461	0.038			-0.397				-0.316
20.35	0.485	0.025			-0.388				-0.322
22.35	0.533	0.022			-0.364				-0.300
23.35	0.556	0.030			-0.369				-0.330
24.35	0.580	0.046			-0.350		-0.390		-0.337
25.35	0.604	0.040			-0.321				-0.307
26.35	0.628	0.052			-0.297				-0.287
27.35	0.652	0.071			-0.299				
30.35	0.723								-0.277
31.35	0.747		0.131	-0.256		-0.225	-0.273	-0.247	-0.236
32.35	0.771		0.141	-0.240		-0.205		-0.217	-0.216
33.35	0.795		0.135	-0.205		-0.190		-0.192	-0.217
34.35	0.818		0.137	-0.188		-0.172		-0.155	-0.198
35.35	0.842		0.146	-0.180		-0.133		-0.150	-0.162
36.35	0.866		0.133	-0.128		-0.119		-0.140	-0.168
37.35	0.890		0.115	-0.109		-0.097		-0.113	-0.135
38.35	0.914		0.100	-0.079		-0.062		-0.113	-0.120
39.35	0.938		0.110	-0.044	-0.045	-0.043		-0.081	-0.102
40.35	0.961		0.073	-0.020	-0.030	-0.022		-0.055	-0.083
41.35	0.985		0.050	0.022	-0.002	-0.011		-0.052	-0.078
42.35	1.009		0.058	0.036	0.027	-0.005		-0.040	-0.106
44.85	1.069		0.025	0.017	0.016	0.005		-0.036	-0.061
45.85	1.092		0.040	0.028	0.022				
46.85	1.116		0.043	0.026	-0.002				

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR							
16.75	0.399			-0.281	-0.224	-0.164		
13.75	0.328		-0.290	-0.296	-0.238	-0.152		
10.75	0.256		-0.324	-0.337	-0.236	-0.168	-0.078	-0.106
7.75	0.185	-0.346	-0.365	-0.390	-0.273	-0.153	-0.070	-0.053
6.75	0.161			-0.247	-0.140	-0.052	-0.040	-0.036
5.75	0.137	-0.377	-0.396	-0.400	-0.279	-0.161	-0.054	-0.050
4.75	0.113			-0.267	-0.155	-0.043	-0.040	-0.024
4.25	0.101	-0.409	-0.405	-0.350				
3.75	0.089			-0.225	-0.119	-0.011	-0.005	0.005
2.75	0.066			-0.268	-0.144	-0.028	-0.023	-0.004
1.75	0.042			-0.275	-0.177	-0.020	0.008	0.006
0.75	0.018			-0.256	-0.128	-0.002	0.027	0.016
-0.25	-0.006					0.022	0.036	0.017
-1.25	-0.030			0.131	0.133	0.050	0.058	0.025
-2.25	-0.054			0.118	0.124	0.071	0.080	0.051
-2.75	-0.066		0.041	0.046				
-3.25	-0.077				0.121	0.114	0.074	0.080	0.030
-4.25	-0.101	0.093	0.045	0.039	0.108	0.113	0.068	0.080	0.049
-5.25	-0.125				0.097	0.096	0.077	0.077	0.038
-6.25	-0.149	0.091	0.058	0.045	0.085	0.091	0.077		-0.010
-9.25	-0.220		0.072	0.048	0.072	0.095	0.072		
-12.25	-0.292		0.062	0.057	0.078	0.084			
-15.25	-0.363			0.058	0.068	0.064			

RUN.SEQ
171.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.829	4.382	9.97	2289	1459	555.4	701.7	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.445	0.084	0.529	-0.0222	-0.0133	-0.0356	0.2280	29.99	40.92	31.72	43.07	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.348	0.100	0.447	-0.0444	-0.0199	-0.0643	37.78	44.94	39.38	0.641	0.137
0.296	0.436	0.086	0.522	-0.0339	-0.0222	-0.0560	32.77	50.80	35.74	0.637	-0.000
0.500	0.509	0.095	0.604	-0.0289	-0.0286	-0.0575	30.69	55.16	34.53	0.604	-0.109
0.697	0.553	0.079	0.632	-0.0296	-0.0252	-0.0549	30.35	57.02	33.68	0.497	-0.140
0.894	0.565	0.027	0.592	-0.0477	-0.0172	-0.0649	33.43	88.44	35.95	0.340	-0.110

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.544	0.544	0.544	0.544	0.544
0.01			0.406		
0.02			0.331		
0.03			0.283		
0.04			0.249		
0.05	0.233	0.206	0.205	0.190	0.092
0.10	0.154	0.120	0.112	0.093	0.014
0.15	0.113	0.079	0.072	0.046	-0.027
0.20	0.099	0.057	0.039	0.028	-0.049
0.30	0.057	0.010	0.003	-0.012	-0.091
0.40	0.027	-0.008	0.000	-0.027	-0.083
0.50	-0.003	-0.026	-0.009	-0.047	-0.076
0.55					
0.60	0.030	0.035	0.063	0.028	-0.008
0.65			0.119		
0.70	0.116	0.136	0.157	0.135	0.114
0.75	0.153	0.167	0.191	0.187	0.148
0.80	0.167	0.188	0.202	0.200	0.168
0.85	0.165	0.185	0.199	0.199	0.160
0.90	0.141	0.151	0.172	0.181	0.140
0.95	0.090	0.090	0.122	0.121	0.094
1.00	0.024	0.035	0.052	0.048	-0.024

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.470	0.381	0.289	0.241	0.260
0.003			-0.105		
0.006			-0.319	-0.358	
0.01	-0.239	-0.466	-0.573	-0.612	-0.517
0.02	-0.545	-0.830	-0.905	-0.888	-0.738
0.03	-0.636	-0.984	-1.067	-1.023	-0.824
0.04	-0.627	-1.079	-1.132	-1.026	-0.779
0.05	-0.636	-1.085	-1.177	-1.163	-0.947
0.06	-0.602	-1.046	-1.199	-1.179	-0.964
0.08	-0.569	-1.094	-1.188	-1.145	-0.954
0.10	-0.550	-0.886	-1.225	-1.153	-0.958
0.125	-0.525	-0.823	-1.222	-1.182	-0.974
0.15	-0.514	-0.777	-1.208	-1.209	-0.983
0.175	-0.504	-0.711	-0.976	-1.194	-0.990
0.20	-0.500	-0.664	-0.787	-1.197	-1.006
0.225	-0.495	-0.618	-0.719	-1.180	-1.008
0.25	-0.471	-0.580	-0.684	-0.941	-1.009
0.30	-0.453	-0.547	-0.639	-0.703	-1.087
0.35	-0.441	-0.516	-0.608	-0.683	-1.153
0.40	-0.443	-0.493	-0.598	-0.675	-0.813
0.45	-0.407	-0.473	-0.561	-0.639	-0.562
0.50	-0.386	-0.470	-0.536	-0.568	-0.437
0.55	-0.373	-0.422	-0.453	-0.461	-0.398
0.60	-0.349	-0.377	-0.375	-0.364	-0.351
0.65	-0.319	-0.321	-0.316	-0.299	-0.300
0.70	-0.297	-0.263	-0.269	-0.251	-0.260
0.75	-0.254	-0.229	-0.198	-0.207	-0.238
0.80	-0.200	-0.184	-0.166	-0.163	-0.209
0.85	-0.150	-0.124	-0.115	-0.115	-0.175
0.90	-0.091	-0.069	-0.061	-0.040	-0.132
0.95	-0.021	-0.005	0.009	0.011	-0.077
1.00	0.024	0.035	0.052	0.048	-0.024

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.066				-0.417				
11.35	0.270	0.084				-0.422				
12.35	0.294					-0.415		-0.377		
14.35	0.342	0.059				-0.409				
15.35	0.366	0.051				-0.416				-0.343
16.35	0.390	0.049				-0.412				0.181
17.35	0.413	0.040				-0.408		-0.374		-0.315
18.35	0.437	0.023				-0.411				-0.328
19.35	0.461	0.032				-0.409				-0.341
20.35	0.485	0.026				-0.413				-0.344
22.35	0.533	0.023				-0.392				-0.342
23.35	0.556	0.036				-0.388				-0.308
24.35	0.580	0.040				-0.339		-0.355		-0.329
25.35	0.604	0.030				-0.361				-0.309
26.35	0.628	0.054				-0.363				-0.302
27.35	0.652	0.071				-0.343				
30.35	0.723									-0.244
31.35	0.747		0.127		-0.271		-0.253	-0.236	-0.243	-0.227
32.35	0.771		0.139		-0.238		-0.249		-0.232	-0.217
33.35	0.795		0.162		-0.216		-0.242		-0.214	-0.203
34.35	0.818		0.150		-0.198		-0.201		-0.161	-0.199
35.35	0.842		0.143		-0.157		-0.198		-0.161	-0.164
36.35	0.866		0.129		-0.142		-0.159		-0.135	-0.140
37.35	0.890		0.113		-0.115		-0.127		-0.137	-0.139
38.35	0.914		0.090		-0.089		-0.099		-0.127	-0.133
39.35	0.938		0.083	-0.051	-0.042		-0.082		-0.090	-0.117
40.35	0.961		0.058	-0.031	-0.029		-0.066		-0.077	-0.081
41.35	0.985		0.057	0.013	-0.001		-0.040		-0.069	-0.080
42.35	1.009		0.072	0.050	0.024		-0.019		-0.046	-0.114
44.85	1.069		0.041	0.025	0.024		-0.010		-0.040	-0.087
45.85	1.092		0.028	0.020	0.006					
46.85	1.116		0.034	-0.005	-0.006					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.275	-0.195	-0.143			
13.75	0.328		-0.304	-0.278	-0.242	-0.169			
10.75	0.256		-0.315	-0.329	-0.227	-0.140	-0.080	-0.114	
7.75	0.185	-0.377	-0.374	-0.355	-0.236	-0.166	-0.065		-0.048
6.75	0.161				-0.243	-0.135	-0.069	-0.046	-0.040
5.75	0.137	-0.406	-0.403	-0.363	-0.243	-0.151	-0.062	-0.041	-0.039
4.75	0.113				-0.255	-0.148	-0.039	-0.038	-0.028
4.25	0.101	-0.415	-0.408	-0.339					
3.75	0.089				-0.253	-0.159	-0.040	-0.019	-0.010
2.75	0.066				-0.251	-0.142	-0.026	-0.002	0.001
1.75	0.042				-0.272	-0.173	-0.017	-0.006	0.023
0.75	0.018				-0.271	-0.142	-0.001	0.024	0.024
-0.25	-0.006						0.013	0.050	0.025
-1.25	-0.030					0.127	0.129	0.057	0.072	0.041
-2.25	-0.054					0.121	0.132	0.084	0.069	0.040
-2.75	-0.066			0.040	0.040					
-3.25	-0.077					0.100	0.122	0.074	0.070	0.049
-4.25	-0.101	0.077	0.040	0.037	0.098	0.092	0.073	0.068	0.029	
-5.25	-0.125					0.103	0.090	0.080	0.061	0.035
-6.25	-0.149	0.086	0.062	0.037	0.094	0.098	0.066		-0.007	
-9.25	-0.220		0.070	0.052	0.084	0.082	0.058			
-12.25	-0.292		0.062	0.049	0.086	0.080				
-15.25	-0.363			0.046	0.063	0.072				

RUN.SEQ
172.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.822	4.401	10.0	2312	1483	556.3	702.3	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.437	0.085	0.522	-0.0201	-0.0140	-0.0341	0.2236	29.59	41.42	31.53	42.79	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.349	0.097	0.446	-0.0447	-0.0199	-0.0647	37.81	45.49	39.49	0.640	0.136
0.296	0.428	0.090	0.518	-0.0317	-0.0231	-0.0548	32.41	50.64	35.59	0.632	0.001
0.500	0.503	0.090	0.592	-0.0282	-0.0268	-0.0550	30.61	54.91	34.29	0.592	-0.105
0.697	0.543	0.091	0.634	-0.0282	-0.0272	-0.0554	30.20	54.95	33.75	0.499	-0.140
0.894	0.519	0.030	0.549	-0.0442	-0.0169	-0.0611	33.51	81.60	36.13	0.315	-0.103

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.544	0.544	0.544	0.544	0.544
0.01			0.400		
0.02			0.323		
0.03			0.275		
0.04			0.243		
0.05	0.230	0.214	0.207	0.205	0.102
0.10	0.161	0.120	0.112	0.114	0.026
0.15	0.124	0.080	0.063	0.066	-0.009
0.20	0.087	0.057	0.047	0.035	-0.036
0.30	0.050	0.012	0.005	0.001	-0.095
0.40	0.010	-0.000	-0.019	-0.016	-0.087
0.50	-0.002	-0.007	-0.023	-0.016	-0.073
0.55					
0.60	0.030	0.032	0.063	0.050	-0.003
0.65			0.108		
0.70	0.120	0.136	0.141	0.148	0.112
0.75	0.160	0.177	0.183	0.185	0.146
0.80	0.173	0.188	0.196	0.198	0.165
0.85	0.169	0.189	0.196	0.198	0.158
0.90	0.143	0.150	0.172	0.179	0.139
0.95	0.092	0.097	0.120	0.129	0.093
1.00	0.005	0.034	0.048	0.054	-0.014

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.458	0.368	0.274	0.223	0.244
0.003			-0.129		
0.006			-0.346	-0.383	
0.01	-0.261	-0.495	-0.603	-0.642	-0.546
0.02	-0.562	-0.858	-0.937	-0.918	-0.768
0.03	-0.644	-0.998	-1.100	-1.057	-0.853
0.04	-0.642	-1.105	-1.164	-1.068	-0.808
0.05	-0.657	-1.106	-1.212	-1.205	-0.976
0.06	-0.623	-1.061	-1.241	-1.211	-0.989
0.08	-0.589	-1.115	-1.223	-1.182	-0.975
0.10	-0.560	-0.885	-1.247	-1.183	-0.970
0.125	-0.530	-0.814	-1.245	-1.214	-0.978
0.15	-0.521	-0.770	-1.178	-1.235	-0.979
0.175	-0.519	-0.707	-0.816	-1.221	-0.980
0.20	-0.502	-0.653	-0.781	-1.206	-0.979
0.225	-0.485	-0.586	-0.719	-0.951	-0.937
0.25	-0.464	-0.564	-0.684	-0.754	-0.739
0.30	-0.446	-0.548	-0.649	-0.698	-0.729
0.35	-0.416	-0.511	-0.591	-0.683	-0.803
0.40	-0.419	-0.483	-0.562	-0.652	-0.751
0.45	-0.432	-0.449	-0.520	-0.603	-0.649
0.50	-0.397	-0.430	-0.478	-0.540	-0.461
0.55	-0.367	-0.378	-0.426	-0.459	-0.395
0.60	-0.345	-0.346	-0.380	-0.375	-0.351
0.65	-0.315	-0.298	-0.316	-0.311	-0.307
0.70	-0.288	-0.255	-0.266	-0.255	-0.269
0.75	-0.239	-0.221	-0.220	-0.210	-0.242
0.80	-0.198	-0.191	-0.186	-0.162	-0.205
0.85	-0.160	-0.136	-0.123	-0.111	-0.162
0.90	-0.101	-0.072	-0.062	-0.041	-0.115
0.95	-0.036	-0.013	0.006	0.011	-0.070
1.00	0.005	0.034	0.048	0.054	-0.014

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.080				-0.387				
11.35	0.270	0.101				-0.383				
12.35	0.294					-0.398		-0.340		
14.35	0.342	0.067				-0.395				
15.35	0.366	0.068				-0.375				-0.329
16.35	0.390	0.073				-0.396				0.124
17.35	0.413	0.058				-0.402		-0.325		-0.297
18.35	0.437	0.047				-0.391				-0.293
19.35	0.461	0.053				-0.369				-0.294
20.35	0.485	0.046				-0.374				-0.295
22.35	0.533	0.032				-0.346				-0.277
23.35	0.556	0.044				-0.326				-0.283
24.35	0.580	0.069				-0.314		-0.302		-0.277
25.35	0.604	0.054				-0.328				-0.271
26.35	0.628	0.058				-0.286				-0.240
27.35	0.652	0.073				-0.296				
30.35	0.723									-0.215
31.35	0.747		0.137		-0.241		-0.219	-0.193	-0.216	-0.195
32.35	0.771		0.154		-0.223		-0.195		-0.199	-0.183
33.35	0.795		0.162		-0.201		-0.180		-0.173	-0.177
34.35	0.818		0.150		-0.186		-0.153		-0.166	-0.140
35.35	0.842		0.147		-0.150		-0.153		-0.152	-0.139
36.35	0.866		0.146		-0.108		-0.136		-0.122	-0.142
37.35	0.890		0.125		-0.083		-0.115		-0.121	-0.112
38.35	0.914		0.091		-0.058		-0.084		-0.093	-0.118
39.35	0.938		0.102	-0.020	-0.050		-0.050		-0.078	-0.093
40.35	0.961		0.090	0.004	-0.011		-0.032		-0.048	-0.060
41.35	0.985		0.084	0.049	0.024		-0.012		-0.039	-0.062
42.35	1.009		0.070	0.040	0.041		-0.008		-0.028	-0.105
44.85	1.069		0.040	0.031	0.030		-0.000		-0.029	-0.082
45.85	1.092		0.024	0.027	0.030					
46.85	1.116		0.038	0.020	0.019					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.239	-0.173	-0.126			
13.75	0.328		-0.258	-0.252	-0.176	-0.118			
10.75	0.256		-0.297	-0.277	-0.195	-0.142	-0.062	-0.105	
7.75	0.185	-0.340	-0.325	-0.302	-0.193	-0.125	-0.059		-0.007
6.75	0.161				-0.216	-0.122	-0.039	-0.028	-0.029
5.75	0.137	-0.391	-0.366	-0.303	-0.193	-0.129	-0.034	-0.015	0.006
4.75	0.113				-0.191	-0.112	-0.017	-0.012	-0.007
4.25	0.101	-0.398	-0.402	-0.314					
3.75	0.089				-0.219	-0.136	-0.012	-0.008	-0.000
2.75	0.066				-0.233	-0.105	-0.005	0.020	0.014
1.75	0.042				-0.257	-0.164	0.012	0.030	0.039
0.75	0.018				-0.241	-0.108	0.024	0.041	0.030
-0.25	-0.006						0.049	0.040	0.031
-1.25	-0.030				0.137	0.146	0.084	0.070	0.040
-2.25	-0.054				0.139	0.123	0.089	0.076	0.052
-2.75	-0.066			0.058	0.069				
-3.25	-0.077					0.115	0.121	0.090	0.081	0.061
-4.25	-0.101	0.084	0.061	0.044	0.105	0.117	0.086	0.085	0.062
-5.25	-0.125					0.104	0.106	0.077	0.083	0.059
-6.25	-0.149	0.091	0.067	0.043	0.107	0.102	0.076			0.017
-9.25	-0.220		0.076	0.069	0.090	0.081	0.074			
-12.25	-0.292		0.077	0.078	0.075	0.067				
-15.25	-0.363			0.079	0.063	0.070				

RUN.SEQ
173.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.812	4.384	9.97	2317	1501	556.2	693.4	5.00	17

WING COEFFICIENTS

	CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
	0.427	0.092	0.518	-0.0186	-0.0151	-0.0337	0.2222	29.36	41.49	31.51	42.87	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.341	0.103	0.444	-0.0427	-0.0216	-0.0643	37.53	45.98	39.48	0.636	0.135
0.296	0.419	0.098	0.517	-0.0294	-0.0251	-0.0545	32.02	50.69	35.56	0.630	0.001
0.500	0.476	0.101	0.577	-0.0236	-0.0285	-0.0521	29.95	53.15	34.02	0.577	-0.101
0.697	0.531	0.092	0.622	-0.0287	-0.0279	-0.0566	30.40	55.47	34.09	0.490	-0.139
0.894	0.535	0.035	0.571	-0.0454	-0.0185	-0.0639	33.49	77.20	36.20	0.328	-0.107

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.543	0.543	0.543	0.543	0.543
0.01			0.408		
0.02			0.336		
0.03			0.286		
0.04			0.261		
0.05	0.232	0.219	0.217	0.204	0.101
0.10	0.156	0.130	0.122	0.111	0.031
0.15	0.116	0.087	0.083	0.057	-0.010
0.20	0.090	0.058	0.064	0.030	-0.035
0.30	0.052	0.019	0.023	0.002	-0.084
0.40	0.025	0.012	0.003	-0.009	-0.086
0.50	0.007	0.002	-0.003	-0.015	-0.064
0.55					
0.60	0.045	0.052	0.063	0.052	0.011
0.65			0.114		
0.70	0.132	0.145	0.158	0.144	0.126
0.75	0.166	0.183	0.190	0.187	0.156
0.80	0.178	0.190	0.200	0.203	0.164
0.85	0.172	0.186	0.200	0.202	0.161
0.90	0.139	0.144	0.169	0.178	0.141
0.95	0.090	0.112	0.124	0.133	0.093
1.00	0.006	0.042	0.056	0.054	-0.018

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.449	0.355	0.258	0.209	0.232
0.003			-0.150		
0.006			-0.372	-0.411	
0.01	-0.270	-0.522	-0.633	-0.673	-0.575
0.02	-0.572	-0.882	-0.969	-0.949	-0.793
0.03	-0.654	-1.034	-1.135	-1.092	-0.881
0.04	-0.646	-1.123	-1.195	-1.101	-0.833
0.05	-0.652	-1.113	-1.240	-1.244	-1.009
0.06	-0.626	-1.079	-1.264	-1.234	-1.015
0.08	-0.593	-1.085	-1.242	-1.210	-1.002
0.10	-0.567	-0.887	-1.263	-1.201	-0.998
0.125	-0.540	-0.811	-1.252	-1.234	-0.991
0.15	-0.528	-0.763	-0.941	-1.250	-0.953
0.175	-0.518	-0.682	-0.771	-1.211	-0.962
0.20	-0.477	-0.630	-0.696	-0.998	-0.966
0.225	-0.454	-0.596	-0.652	-0.724	-0.864
0.25	-0.435	-0.555	-0.628	-0.665	-0.793
0.30	-0.445	-0.523	-0.601	-0.660	-0.755
0.35	-0.424	-0.486	-0.545	-0.642	-0.816
0.40	-0.411	-0.464	-0.515	-0.620	-0.855
0.45	-0.396	-0.430	-0.490	-0.596	-0.804
0.50	-0.367	-0.390	-0.447	-0.550	-0.449
0.55	-0.342	-0.371	-0.408	-0.470	-0.375
0.60	-0.326	-0.343	-0.366	-0.382	-0.349
0.65	-0.306	-0.306	-0.311	-0.314	-0.315
0.70	-0.287	-0.255	-0.256	-0.255	-0.264
0.75	-0.245	-0.217	-0.215	-0.221	-0.235
0.80	-0.199	-0.182	-0.174	-0.173	-0.201
0.85	-0.151	-0.133	-0.107	-0.113	-0.154
0.90	-0.101	-0.071	-0.037	-0.051	-0.111
0.95	-0.034	-0.000	0.026	0.014	-0.074
1.00	0.006	0.042	0.056	0.054	-0.018

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR								
10.35	0.247	0.071			-0.387				
11.35	0.270	0.088			-0.391				
12.35	0.294				-0.394		-0.365		
14.35	0.342	0.065			-0.389				
15.35	0.366	0.066			-0.374				-0.319
16.35	0.390	0.062			-0.355				0.071
17.35	0.413	0.066			-0.355		-0.345		-0.279
18.35	0.437	0.042			-0.362				-0.276
19.35	0.461	0.035			-0.376				-0.295
20.35	0.485	0.029			-0.364				-0.302
22.35	0.533	0.021			-0.332				-0.290
23.35	0.556	0.026			-0.332				-0.262
24.35	0.580	0.048			-0.318		-0.311		-0.280
25.35	0.604	0.048			-0.316				-0.282
26.35	0.628	0.070			-0.306				-0.259
27.35	0.652	0.071			-0.287				
30.35	0.723								-0.221
31.35	0.747		0.144	-0.210		-0.223	-0.225	-0.216	-0.201
32.35	0.771		0.159	-0.225		-0.205		-0.199	-0.196
33.35	0.795		0.150	-0.189		-0.186		-0.199	-0.178
34.35	0.818		0.144	-0.169		-0.164		-0.178	-0.174
35.35	0.842		0.144	-0.149		-0.142		-0.156	-0.153
36.35	0.866		0.134	-0.138		-0.125		-0.146	-0.137
37.35	0.890		0.134	-0.093		-0.099		-0.103	-0.129
38.35	0.914		0.106	-0.054		-0.091		-0.102	-0.122
39.35	0.938		0.097	-0.049	-0.043	-0.065		-0.086	-0.097
40.35	0.961		0.073	-0.023	-0.013	-0.056		-0.063	-0.101
41.35	0.985		0.077	0.022	0.005	-0.025		-0.056	-0.077
42.35	1.009		0.059	0.047	0.018	-0.006		-0.043	-0.100
44.85	1.069		0.044	0.029	0.029	0.000		-0.019	-0.074
45.85	1.092		0.043	0.025	0.022				
46.85	1.116		0.058	0.040	0.019				

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR							
16.75	0.399			-0.236	-0.194	-0.146		
13.75	0.328		-0.267	-0.262	-0.199	-0.136		
10.75	0.256		-0.279	-0.280	-0.201	-0.137	-0.077	-0.100
7.75	0.185	-0.365	-0.345	-0.311	-0.225	-0.132	-0.053	-0.037
6.75	0.161				-0.216	-0.146	-0.056	-0.043 -0.019
5.75	0.137	-0.392	-0.358	-0.333	-0.228	-0.132	-0.050	-0.030 -0.029
4.75	0.113				-0.224	-0.133	-0.023	-0.014 -0.004
4.25	0.101	-0.394	-0.355	-0.318				
3.75	0.089				-0.223	-0.125	-0.025	-0.006 0.000
2.75	0.066				-0.240	-0.135	-0.008	0.003 0.011
1.75	0.042				-0.248	-0.142	-0.002	0.017 0.034
0.75	0.018				-0.210	-0.138	0.005	0.018 0.029
-0.25	-0.006						0.022	0.047 0.029
-1.25	-0.030				0.144	0.134	0.077	0.059 0.044
-2.25	-0.054				0.122	0.128	0.095	0.071 0.046
-2.75	-0.066			0.066	0.048			
-3.25	-0.077				0.104	0.123	0.078	0.063 0.042
-4.25	-0.101	0.090	0.051	0.045	0.093	0.111	0.066	0.070 0.042
-5.25	-0.125				0.094	0.101	0.052	0.082 0.038
-6.25	-0.149	0.088	0.065	0.062	0.089	0.086	0.061	0.005
-9.25	-0.220		0.079	0.059	0.066	0.083	0.068	
-12.25	-0.292		0.055	0.055	0.075	0.081		
-15.25	-0.363			0.052	0.065	0.073		

RUN.SEQ
174.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.802	4.403	10.0	2339	1531	555.9	689.7	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	C
0.422	0.092	0.514	-0.0186	-0.0143	-0.0329	0.2192	29.41	40.50	31.39	42.67	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.333	0.109	0.442	-0.0389	-0.0226	-0.0616	36.67	45.77	38.91	0.634	0.140
0.296	0.415	0.098	0.513	-0.0302	-0.0254	-0.0556	32.26	50.92	35.83	0.626	-0.001
0.500	0.477	0.102	0.579	-0.0253	-0.0284	-0.0537	30.31	52.78	34.27	0.579	-0.103
0.697	0.523	0.085	0.608	-0.0305	-0.0269	-0.0574	30.82	56.61	34.43	0.479	-0.137
0.894	0.527	0.027	0.554	-0.0480	-0.0154	-0.0634	34.11	81.89	36.45	0.318	-0.104

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.542	0.542	0.542	0.542	0.542
0.01			0.417		
0.02			0.340		
0.03			0.288		
0.04			0.254		
0.05	0.240	0.201	0.216	0.198	0.105
0.10	0.167	0.134	0.125	0.092	0.027
0.15	0.128	0.094	0.094	0.040	-0.019
0.20	0.091	0.066	0.067	0.033	-0.042
0.30	0.057	0.026	0.028	-0.008	-0.079
0.40	0.031	0.008	0.002	-0.031	-0.092
0.50	0.022	-0.008	-0.012	-0.024	-0.076
0.55					
0.60	0.065	0.055	0.064	0.049	-0.007
0.65			0.120		
0.70	0.140	0.145	0.160	0.159	0.104
0.75	0.161	0.185	0.191	0.182	0.137
0.80	0.172	0.189	0.200	0.197	0.148
0.85	0.174	0.194	0.198	0.195	0.158
0.90	0.135	0.158	0.174	0.176	0.139
0.95	0.084	0.107	0.128	0.129	0.086
1.00	0.031	0.032	0.043	0.038	-0.021

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.427	0.338	0.242	0.188	0.217
0.003			-0.174		
0.006			-0.396	-0.442	
0.01	-0.298	-0.546	-0.662	-0.708	-0.603
0.02	-0.593	-0.915	-1.006	-0.987	-0.826
0.03	-0.663	-1.070	-1.171	-1.128	-0.911
0.04	-0.655	-1.155	-1.238	-1.134	-0.859
0.05	-0.658	-1.102	-1.275	-1.284	-1.037
0.06	-0.626	-1.102	-1.302	-1.270	-1.049
0.08	-0.593	-1.000	-1.275	-1.240	-1.027
0.10	-0.569	-0.874	-1.291	-1.227	-1.013
0.125	-0.533	-0.795	-1.239	-1.261	-1.000
0.15	-0.520	-0.742	-0.831	-1.270	-0.949
0.175	-0.507	-0.670	-0.734	-0.933	-0.785
0.20	-0.488	-0.623	-0.686	-0.718	-0.627
0.225	-0.464	-0.591	-0.648	-0.669	-0.611
0.25	-0.441	-0.549	-0.615	-0.655	-0.658
0.30	-0.438	-0.504	-0.591	-0.672	-0.822
0.35	-0.410	-0.481	-0.553	-0.653	-0.916
0.40	-0.396	-0.448	-0.518	-0.626	-0.943
0.45	-0.389	-0.407	-0.469	-0.592	-0.700
0.50	-0.358	-0.399	-0.438	-0.535	-0.451
0.55	-0.333	-0.365	-0.406	-0.456	-0.402
0.60	-0.308	-0.331	-0.368	-0.368	-0.362
0.65	-0.283	-0.305	-0.313	-0.315	-0.322
0.70	-0.274	-0.252	-0.265	-0.264	-0.275
0.75	-0.244	-0.218	-0.207	-0.228	-0.246
0.80	-0.197	-0.193	-0.177	-0.188	-0.202
0.85	-0.147	-0.143	-0.124	-0.130	-0.165
0.90	-0.081	-0.078	-0.059	-0.057	-0.124
0.95	-0.012	-0.010	0.006	-0.001	-0.070
1.00	0.031	0.032	0.043	0.038	-0.021

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.069				-0.384				
11.35	0.270	0.079				-0.390				
12.35	0.294					-0.377		-0.335		
14.35	0.342	0.075				-0.377				
15.35	0.366	0.061				-0.369				-0.320
16.35	0.390	0.052				-0.361				0.022
17.35	0.413	0.047				-0.368		-0.326		-0.277
18.35	0.437	0.027				-0.357				-0.277
19.35	0.461	0.041				-0.345				-0.284
20.35	0.485	0.027				-0.333				-0.290
22.35	0.533	0.040				-0.305				-0.272
23.35	0.556	0.048				-0.325				-0.255
24.35	0.580	0.053				-0.320		-0.281		-0.251
25.35	0.604	0.046				-0.306				-0.229
26.35	0.628	0.061				-0.265				-0.230
27.35	0.652	0.082				-0.262				
30.35	0.723									-0.187
31.35	0.747		0.151		-0.230		-0.183	-0.201	-0.192	-0.193
32.35	0.771		0.158		-0.199		-0.189		-0.187	-0.176
33.35	0.795		0.164		-0.170		-0.186		-0.192	-0.163
34.35	0.818		0.147		-0.165		-0.160		-0.163	-0.147
35.35	0.842		0.149		-0.139		-0.149		-0.155	-0.130
36.35	0.866		0.142		-0.108		-0.132		-0.129	-0.118
37.35	0.890		0.124		-0.073		-0.101		-0.113	-0.109
38.35	0.914		0.097		-0.052		-0.081		-0.089	-0.099
39.35	0.938		0.095	-0.022	-0.024		-0.070		-0.064	-0.089
40.35	0.961		0.080	-0.000	0.012		-0.035		-0.046	-0.079
41.35	0.985		0.077	0.042	0.045		-0.011		-0.042	-0.075
42.35	1.009		0.080	0.052	0.044		0.005		-0.034	-0.089
44.85	1.069		0.042	0.034	0.048		0.007		-0.033	-0.053
45.85	1.092		0.039	0.027	0.032					
46.85	1.116		0.038	0.017	0.026					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.210	-0.160	-0.136			
13.75	0.328		-0.253	-0.221	-0.168	-0.141			
10.75	0.256		-0.277	-0.251	-0.193	-0.118	-0.075	-0.089	
7.75	0.185	-0.335	-0.326	-0.281	-0.201	-0.127	-0.047		-0.040
6.75	0.161				-0.192	-0.129	-0.042	-0.034	-0.033
5.75	0.137	-0.365	-0.348	-0.291	-0.206	-0.116	-0.031	-0.009	-0.002
4.75	0.113				-0.202	-0.123	-0.021	-0.016	-0.008
4.25	0.101	-0.377	-0.368	-0.320					
3.75	0.089				-0.183	-0.132	-0.011	0.005	0.007
2.75	0.066				-0.222	-0.121	-0.007	-0.012	0.002
1.75	0.042				-0.204	-0.166	0.008	0.021	0.020
0.75	0.018				-0.230	-0.108	0.045	0.044	0.048
-0.25	-0.006						0.042	0.052	0.034
-1.25	-0.030					0.151	0.142	0.077	0.080	0.042
-2.25	-0.054					0.139	0.133	0.084	0.082	0.053
-2.75	-0.066		0.047	0.053						
-3.25	-0.077						0.123	0.131	0.077	0.074	0.058
-4.25	-0.101	0.079	0.073	0.055	0.115	0.125	0.077	0.097	0.068
-5.25	-0.125					0.120	0.120	0.083	0.083	0.052
-6.25	-0.149	0.098	0.089	0.073	0.107	0.124	0.086		0.021
-9.25	-0.220		0.095	0.061	0.096	0.078	0.089		
-12.25	-0.292		0.082	0.063	0.096	0.073			
-15.25	-0.363			0.071	0.068	0.080			

RUN.SEQ
178.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.859	3.965	9.02	2027	1252	553.2	647.0	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.453	0.093	0.546	-0.0285	-0.0145	-0.0431	0.2358	31.30	40.58	32.89	43.17	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.344	0.115	0.459	-0.0465	-0.0244	-0.0709	38.52	46.22	40.45	0.657	0.130
0.296	0.428	0.110	0.538	-0.0350	-0.0286	-0.0636	33.18	50.92	36.82	0.657	-0.009
0.500	0.523	0.099	0.621	-0.0349	-0.0307	-0.0656	31.68	56.11	35.56	0.621	-0.118
0.697	0.590	0.080	0.670	-0.0373	-0.0275	-0.0648	31.33	59.32	34.68	0.527	-0.152
0.894	0.588	0.006	0.594	-0.0643	-0.0131	-0.0774	35.93	235.5	38.03	0.341	-0.115

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.548	0.548	0.548	0.548	0.548
0.01			0.394		
0.02			0.320		
0.03			0.272		
0.04			0.243		
0.05	0.242	0.236	0.218	0.194	0.059
0.10	0.169	0.142	0.120	0.103	0.004
0.15	0.134	0.110	0.058	0.050	-0.041
0.20	0.104	0.080	0.045	0.003	-0.062
0.30	0.076	0.034	0.011	-0.029	-0.135
0.40	0.030	0.010	-0.011	-0.041	-0.123
0.50	0.026	-0.011	-0.021	-0.035	-0.105
0.55					
0.60	0.048	0.065	0.058	0.043	-0.018
0.65			0.124		
0.70	0.128	0.175	0.174	0.155	0.098
0.75	0.177	0.197	0.201	0.186	0.135
0.80	0.197	0.201	0.217	0.200	0.156
0.85	0.185	0.205	0.217	0.209	0.157
0.90	0.150	0.167	0.186	0.182	0.128
0.95	0.105	0.124	0.138	0.137	0.083
1.00	0.026	0.072	0.054	0.061	-0.069

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.499	0.417	0.334	0.288	0.303
0.003			-0.050		
0.006			-0.258	-0.280	
0.01	-0.186	-0.403	-0.496	-0.516	-0.440
0.02	-0.478	-0.751	-0.813	-0.795	-0.649
0.03	-0.594	-0.800	-0.972	-0.929	-0.736
0.04	-0.582	-0.996	-1.033	-0.945	-0.706
0.05	-0.606	-1.023	-1.099	-1.085	-0.881
0.06	-0.571	-1.000	-1.113	-1.083	-0.893
0.08	-0.542	-1.010	-1.107	-1.063	-0.875
0.10	-0.517	-0.899	-1.135	-1.087	-0.880
0.125	-0.491	-0.826	-1.133	-1.097	-0.903
0.15	-0.480	-0.752	-1.135	-1.131	-0.917
0.175	-0.482	-0.692	-1.108	-1.113	-0.937
0.20	-0.479	-0.642	-0.873	-1.126	-0.969
0.225	-0.463	-0.603	-0.742	-1.119	-0.987
0.25	-0.445	-0.570	-0.687	-1.113	-0.990
0.30	-0.444	-0.553	-0.662	-1.090	-1.055
0.35	-0.428	-0.506	-0.634	-0.778	-1.121
0.40	-0.428	-0.496	-0.644	-0.754	-0.665
0.45	-0.435	-0.485	-0.629	-0.745	-0.650
0.50	-0.423	-0.491	-0.612	-0.731	-0.632
0.55	-0.404	-0.455	-0.585	-0.661	-0.608
0.60	-0.390	-0.390	-0.526	-0.436	-0.560
0.65	-0.376	-0.329	-0.324	-0.262	-0.483
0.70	-0.326	-0.237	-0.245	-0.215	-0.351
0.75	-0.233	-0.195	-0.184	-0.191	-0.285
0.80	-0.182	-0.163	-0.147	-0.144	-0.196
0.85	-0.146	-0.108	-0.091	-0.098	-0.147
0.90	-0.072	-0.143	-0.040	-0.021	-0.109
0.95	-0.001	0.022	0.011	0.025	-0.078
1.00	0.026	0.072	0.054	0.061	-0.069

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.077				-0.388				
11.35	0.270	0.112				-0.397				
12.35	0.294					-0.407		-0.345		
14.35	0.342	0.088				-0.410				
15.35	0.366	0.072				-0.408				-0.350
16.35	0.390	0.067				-0.407				0.415
17.35	0.413	0.054				-0.412		-0.378		-0.319
18.35	0.437	0.050				-0.421				-0.324
19.35	0.461	0.054				-0.434				-0.337
20.35	0.485	0.044				-0.426				-0.333
22.35	0.533	0.044				-0.411				-0.317
23.35	0.556	0.052				-0.427				-0.340
24.35	0.580	0.058				-0.414		-0.369		-0.340
25.35	0.604	0.058				-0.406				-0.327
26.35	0.628	0.063				-0.396				-0.361
27.35	0.652	0.083				-0.349				
30.35	0.723									-0.290
31.35	0.747		0.143		-0.284		-0.253	-0.224	-0.304	-0.262
32.35	0.771		0.151		-0.236		-0.258		-0.252	-0.257
33.35	0.795		0.161		-0.244		-0.253		-0.204	-0.237
34.35	0.818		0.156		-0.216		-0.195		-0.185	-0.168
35.35	0.842		0.155		-0.178		-0.159		-0.174	-0.147
36.35	0.866		0.150		-0.106		-0.143		-0.141	-0.131
37.35	0.890		0.132		-0.099		-0.111		-0.130	-0.109
38.35	0.914		0.115		-0.084		-0.075		-0.093	-0.122
39.35	0.938		0.121	-0.013	-0.048		-0.077		-0.089	-0.099
40.35	0.961		0.100	0.011	-0.025		-0.056		-0.052	-0.091
41.35	0.985		0.070	0.049	0.001		-0.012		-0.046	-0.073
42.35	1.009		0.079	0.046	0.027		0.003		-0.035	-0.111
44.85	1.069		0.050	0.030	0.029		-0.004		-0.035	-0.061
45.85	1.092		0.040	0.025	0.018					
46.85	1.116		0.051	0.022	0.015					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.301	-0.224	-0.145			
13.75	0.328		-0.289	-0.299	-0.228	-0.159			
10.75	0.256		-0.319	-0.340	-0.262	-0.131	-0.073	-0.111	
7.75	0.185	-0.345	-0.378	-0.369	-0.224	-0.147	-0.027		-0.019
6.75	0.161				-0.304	-0.141	-0.046	-0.035	-0.035
5.75	0.137	-0.378	-0.394	-0.415	-0.239	-0.143	-0.034	-0.027	-0.020
4.75	0.113				-0.266	-0.138	-0.029	-0.022	-0.006
4.25	0.101	-0.407	-0.412	-0.414					
3.75	0.089				-0.253	-0.143	-0.012	0.003	-0.004
2.75	0.066				-0.266	-0.119	0.002	0.014	0.015
1.75	0.042				-0.288	-0.127	0.027	0.032	0.036
0.75	0.018				-0.284	-0.106	0.001	0.027	0.029
-0.25	-0.006						0.049	0.046	0.030
-1.25	-0.030					0.143	0.150	0.070	0.079	0.050
-2.25	-0.054					0.129	0.142	0.088	0.095	0.062
-2.75	-0.066			0.054	0.058					
-3.25	-0.077					0.120	0.135	0.086	0.087	0.059
-4.25	-0.101	0.096	0.057	0.040	0.121	0.125	0.095	0.083	0.069
-5.25	-0.125				0.110	0.113	0.068	0.082	0.073
-6.25	-0.149	0.099	0.068	0.046	0.102	0.118	0.075		0.013
-9.25	-0.220		0.090	0.071	0.093	0.091	0.075		
-12.25	-0.292		0.076	0.072	0 089	0.088			
-15.25	-0.363			0.071	.079	0.074			

RUN.SEQ
179.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.821	3.995	9.09	2083	1338	552.6	630.9	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.438	0.084	0.522	-0.0213	-0.0133	-0.0346	0.2240	29.86	40.78	31.62	42.90	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.348	0.097	0.445	-0.0444	-0.0193	-0.0637	37.77	44.97	39.34	0.637	0.137
0.296	0.423	0.097	0.520	-0.0319	-0.0255	-0.0573	32.53	51.28	36.02	0.635	-0.002
0.500	0.496	0.090	0.586	-0.0281	-0.0264	-0.0545	30.67	54.38	34.31	0.585	-0.104
0.697	0.551	0.080	0.632	-0.0303	-0.0251	-0.0554	30.50	56.23	33.77	0.497	-0.140
0.894	0.547	0.017	0.564	-0.0462	-0.0147	-0.0609	33.44	112.4	35.79	0.324	-0.105

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.544	0.544	0.544	0.544	0.544
0.01			0.402		
0.02			0.336		
0.03			0.282		
0.04			0.246		
0.05	0.221	0.211	0.217	0.189	0.077
0.10	0.157	0.116	0.111	0.100	0.020
0.15	0.129	0.079	0.066	0.047	-0.021
0.20	0.096	0.066	0.041	0.031	-0.054
0.30	0.053	0.022	0.005	-0.007	-0.116
0.40	0.001	0.007	-0.006	-0.028	-0.106
0.50	0.009	-0.010	-0.027	-0.045	-0.085
0.55					
0.60	0.032	0.052	0.049	0.036	-0.016
0.65			0.104		
0.70	0.118	0.148	0.153	0.142	0.104
0.75	0.155	0.182	0.187	0.185	0.134
0.80	0.167	0.201	0.193	0.202	0.160
0.85	0.166	0.191	0.195	0.199	0.156
0.90	0.131	0.152	0.168	0.165	0.127
0.95	0.086	0.105	0.125	0.119	0.090
1.00	0.013	0.043	0.032	0.043	-0.031

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.453	0.364	0.274	0.224	0.242
0.005			-0.134		
0.006			-0.353	-0.386	
0.01	-0.255	-0.499	-0.610	-0.646	-0.547
0.02	-0.550	-0.855	-0.941	-0.925	-0.769
0.03	-0.658	-0.971	-1.109	-1.059	-0.859
0.04	-0.644	-1.110	-1.167	-1.080	-0.826
0.05	-0.646	-1.125	-1.236	-1.245	-1.000
0.06	-0.612	-1.065	-1.241	-1.217	-0.995
0.08	-0.580	-1.098	-1.218	-1.195	-0.983
0.10	-0.553	-0.863	-1.239	-1.206	-0.986
0.125	-0.537	-0.829	-1.245	-1.209	-0.993
0.15	-0.510	-0.755	-1.146	-1.241	-0.998
0.175	-0.493	-0.697	-0.738	-1.212	-1.004
0.20	-0.496	-0.630	-0.729	-1.217	-1.022
0.225	-0.469	-0.592	-0.681	-1.028	-1.008
0.25	-0.451	-0.549	-0.650	-0.774	-1.013
0.30	-0.451	-0.516	-0.638	-0.703	-0.916
0.35	-0.450	-0.508	-0.594	-0.681	-0.796
0.40	-0.443	-0.481	-0.564	-0.647	-0.723
0.45	-0.418	-0.431	-0.519	-0.601	-0.615
0.50	-0.397	-0.417	-0.467	-0.546	-0.454
0.55	-0.356	-0.382	-0.416	-0.478	-0.399
0.60	-0.346	-0.350	-0.367	-0.381	-0.361
0.65	-0.322	-0.301	-0.310	-0.319	-0.335
0.70	-0.288	-0.260	-0.273	-0.272	-0.284
0.75	-0.241	-0.230	-0.210	-0.213	-0.256
0.80	-0.196	-0.192	-0.174	-0.169	-0.205
0.85	-0.146	-0.151	-0.119	-0.120	-0.158
0.90	-0.092	-0.068	-0.067	-0.052	-0.130
0.95	-0.038	-0.007	-0.006	0.002	-0.083
1.00	0.013	0.043	0.032	0.043	-0.031

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.061				-0.406				
11.35	0.270	0.093				-0.409				
12.35	0.294					-0.406		-0.368		
14.35	0.342	0.068				-0.417				
15.35	0.366	0.065				-0.418				-0.338
16.35	0.390	0.052				-0.394				0.244
17.35	0.413	0.052				-0.407		-0.388		-0.334
18.35	0.437	0.045				-0.405				-0.335
19.35	0.461	0.043				-0.385				-0.314
20.35	0.485	0.041				-0.369				-0.319
22.35	0.533	0.030				-0.343				-0.311
23.35	0.556	0.042				-0.375				-0.317
24.35	0.580	0.055				-0.358		-0.323		-0.315
25.35	0.604	0.054				-0.357				-0.307
26.35	0.628	0.054				-0.312				-0.314
27.35	0.652	0.070				-0.294				
30.35	0.723									-0.256
31.35	0.747		0.131		-0.237		-0.243	-0.248	-0.233	-0.233
32.35	0.771		0.145		-0.226		-0.234		-0.217	-0.222
33.35	0.795		0.149		-0.229		-0.216		-0.199	-0.215
34.35	0.818		0.146		-0.186		-0.206		-0.209	-0.166
35.35	0.842		0.142		-0.145		-0.144		-0.176	-0.162
36.35	0.866		0.134		-0.144		-0.125		-0.137	-0.157
37.35	0.890		0.115		-0.111		-0.124		-0.118	-0.147
38.35	0.914		0.091		-0.074		-0.095		-0.118	-0.139
39.35	0.938		0.103	-0.056	-0.039		-0.077		-0.117	-0.115
40.35	0.961		0.090	-0.025	-0.027		-0.053		-0.076	-0.105
41.35	0.985		0.082	0.018	-0.002		-0.042		-0.057	-0.092
42.35	1.009		0.065	0.031	0.009		-0.024		-0.051	-0.122
44.85	1.069		0.033	0.028	0.012		-0.024		-0.037	-0.089
45.85	1.092		0.037	0.031	0.011					
46.85	1.116		0.039	0.009	0.002					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.281	-0.227	-0.163			
13.75	0.328		-0.297	-0.277	-0.230	-0.164			
10.75	0.256		-0.334	-0.315	-0.233	-0.157	-0.092	-0.122	
7.75	0.185	-0.368	-0.388	-0.323	-0.248	-0.150	-0.084		-0.073
6.75	0.161				-0.233	-0.137	-0.057	-0.051	-0.037
5.75	0.137	-0.405	-0.406	-0.370	-0.261	-0.154	-0.045	-0.035	-0.061
4.75	0.113				-0.265	-0.156	-0.061	-0.045	-0.032
4.25	0.101	-0.406	-0.407	-0.358					
3.75	0.089				-0.243	-0.125	-0.042	-0.024	-0.024
2.75	0.066				-0.269	-0.148	-0.037	-0.025	-0.008
1.75	0.042				-0.289	-0.174	-0.019	0.013	0.017
0.75	0.018				-0.237	-0.144	-0.002	0.009	0.012
-0.25	-0.006						0.018	0.031	0.028
-1.25	-0.030					0.131	0.134	0.082	0.065	0.033
-2.25	-0.054					0.134	0.113	0.069	0.072	0.042
-2.75	-0.066			0.052	0.055					
-3.25	-0.077					0.123	0.117	0.065	0.057	0.037
-4.25	-0.101	0.085	0.039	0.034	0.115	0.092	0.068	0.058	0.032
-5.25	-0.125				0.107	0.088	0.061	0.055	0.029
-6.25	-0.149	0.086	0.058	0.043	0.088	0.091	0.053		-0.030
-9.25	-0.220		0.070	0.048	0.073	0.070	0.062		
-12.25	-0.292		0.065	0.055	0.067	0.062			
-15.25	-0.363			0.048	0.055	0.060			

RUN.SEQ
180.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.821	3.495	7.95	1814	1165	550.6	549.8	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.437	0.087	0.524	-0.0196	-0.0135	-0.0330	0.2241	29.47	40.52	31.30	42.77	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.349	0.099	0.448	-0.0442	-0.0188	-0.0630	37.66	44.03	39.07	0.642	0.140
0.296	0.428	0.098	0.526	-0.0313	-0.0243	-0.0556	32.30	49.93	35.58	0.641	0.001
0.500	0.498	0.093	0.592	-0.0269	-0.0281	-0.0550	30.40	55.10	34.31	0.591	-0.105
0.697	0.534	0.084	0.618	-0.0269	-0.0264	-0.0533	30.03	56.47	33.62	0.487	-0.136
0.894	0.546	0.027	0.573	-0.0448	-0.0172	-0.0620	33.20	88.47	35.82	0.329	-0.106

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.544	0.544	0.544	0.544	0.544
0.01			0.419		
0.02			0.333		
0.03			0.274		
0.04			0.239		
0.05	0.221	0.214	0.210	0.198	0.116
0.10	0.157	0.156	0.111	0.102	0.008
0.15	0.119	0.101	0.069	0.055	-0.030
0.20	0.105	0.068	0.042	0.023	-0.052
0.30	0.060	0.017	0.002	-0.005	-0.103
0.40	0.022	0.005	-0.003	-0.025	-0.084
0.50	0.006	-0.012	-0.021	-0.037	-0.079
0.55					
0.60	0.040	0.047	0.065	0.041	-0.006
0.65			0.108		
0.70	0.114	0.148	0.153	0.147	0.114
0.75	0.134	0.193	0.186	0.184	0.155
0.80	0.153	0.182	0.199	0.203	0.162
0.85	0.162	0.190	0.198	0.206	0.165
0.90	0.128	0.161	0.179	0.171	0.143
0.95	0.082	0.097	0.131	0.123	0.091
1.00	0.025	0.034	0.048	0.059	-0.010

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.460	0.370	0.273	0.225	0.244
0.003			-0.134		
0.006			-0.356	-0.386	
0.01	-0.266	-0.498	-0.613	-0.642	-0.555
0.02	-0.561	-0.859	-0.944	-0.918	-0.776
0.03	-0.649	-0.982	-1.111	-1.062	-0.864
0.04	-0.644	-1.099	-1.156	-1.082	-0.836
0.05	-0.647	-1.145	-1.242	-1.255	-1.009
0.06	-0.616	-1.078	-1.252	-1.220	-0.985
0.08	-0.578	-1.085	-1.232	-1.187	-0.971
0.10	-0.559	-0.885	-1.248	-1.198	-0.978
0.125	-0.534	-0.806	-1.240	-1.212	-0.982
0.15	-0.524	-0.780	-1.214	-1.242	-0.989
0.175	-0.522	-0.726	-0.851	-1.217	-0.993
0.20	-0.503	-0.665	-0.748	-1.208	-1.007
0.225	-0.488	-0.618	-0.686	-0.909	-1.002
0.25	-0.467	-0.570	-0.644	-0.739	-1.012
0.30	-0.465	-0.530	-0.629	-0.660	-0.989
0.35	-0.428	-0.516	-0.585	-0.632	-0.783
0.40	-0.429	-0.477	-0.550	-0.615	-0.714
0.45	-0.412	-0.434	-0.526	-0.564	-0.610
0.50	-0.381	-0.422	-0.470	-0.527	-0.465
0.55	-0.372	-0.393	-0.397	-0.463	-0.416
0.60	-0.338	-0.361	-0.378	-0.377	-0.369
0.65	-0.317	-0.315	-0.332	-0.310	-0.315
0.70	-0.292	-0.267	-0.277	-0.261	-0.268
0.75	-0.240	-0.218	-0.214	-0.222	-0.245
0.80	-0.210	-0.171	-0.173	-0.167	-0.209
0.85	-0.155	-0.119	-0.111	-0.110	-0.156
0.90	-0.109	-0.077	-0.059	-0.038	-0.115
0.95	-0.023	-0.009	0.006	0.022	-0.076
1.00	0.025	0.034	0.048	0.059	-0.010

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.037				-0.396				
11.35	0.270	0.087				-0.390				
12.35	0.294					-0.409		-0.343		
14.35	0.342	0.085				-0.406				
15.35	0.366	0.079				-0.380				-0.337
16.35	0.390	0.062				-0.384				0.511
17.35	0.413	0.053				-0.379		-0.340		-0.293
18.35	0.437	0.053				-0.376				-0.304
19.35	0.461	0.047				-0.375				-0.293
20.35	0.485	0.044				-0.373				-0.300
22.35	0.533	0.047				-0.347				-0.299
23.35	0.556	0.044				-0.338				-0.284
24.35	0.580	0.058				-0.338		-0.330		-0.273
25.35	0.604	0.062				-0.336				-0.275
26.35	0.628	0.062				-0.302				-0.269
27.35	0.652	0.094				-0.298				
30.35	0.723									-0.222
31.35	0.747		0.162		-0.244		-0.226	-0.223	-0.232	-0.217
32.35	0.771		0.161		-0.204		-0.199		-0.199	-0.187
33.35	0.795		0.152		-0.199		-0.203		-0.178	-0.182
34.35	0.818		0.148		-0.158		-0.159		-0.167	-0.173
35.35	0.842		0.139		-0.116		-0.154		-0.149	-0.150
36.35	0.866		0.136		-0.132		-0.133		-0.126	-0.132
37.35	0.890		0.130		-0.099		-0.109		-0.115	-0.128
38.35	0.914		0.103		-0.061		-0.089		-0.094	-0.110
39.35	0.938		0.095	-0.044	-0.035		-0.059		-0.071	-0.098
40.35	0.961		0.081	-0.002	-0.019		-0.042		-0.052	-0.088
41.35	0.985		0.059	0.043	0.006		-0.029		-0.041	-0.075
42.35	1.009		0.086	0.054	0.014		-0.018		-0.042	-0.132
44.85	1.069		0.049	0.040	0.012		0.001		-0.019	-0.062
45.85	1.092		0.038	0.033	0.015					
46.85	1.116		0.041	0.021	0.018					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.269	-0.189	-0.146			
13.75	0.328		-0.250	-0.270	-0.205	-0.159			
10.75	0.256		-0.293	-0.273	-0.217	-0.132	-0.075	-0.132	
7.75	0.185	-0.343	-0.340	-0.330	-0.223	-0.145	-0.052		-0.026
6.75	0.161				-0.232	-0.126	-0.041	-0.042	-0.019
5.75	0.137	-0.377	-0.381	-0.354	-0.207	-0.131	-0.053	-0.011	-0.018
4.75	0.113				-0.196	-0.135	-0.023	0.002	-0.017
4.25	0.101	-0.409	-0.379	-0.338					
3.75	0.089				-0.226	-0.133	-0.029	-0.018	0.001
2.75	0.066				-0.235	-0.133	0.010	0.004	-0.014
1.75	0.042				-0.251	-0.104	0.022	0.020	0.009
0.75	0.018				-0.244	-0.132	0.006	0.014	0.012
-0.25	-0.006						0.043	0.054	0.040
-1.25	-0.030				0.162	0.136	0.059	0.086	0.049
-2.25	-0.054				0.131	0.130	0.076	0.092	0.056
-2.75	-0.066		0.053	0.058					
-3.25	-0.077				0.123	0.123	0.074	0.079	0.055
-4.25	-0.101	0.078	0.054	0.033	0.118	0.117	0.066	0.074	0.057
-5.25	-0.125				0.108	0.101	0.087	0.074	0.052
-6.25	-0.149	0.080	0.069	0.046	0.102	0.110	0.091		-0.018
-9.25	-0.220		0.073	0.054	0.079	0.096	0.092		
-12.25	-0.292		0.063	0.057	0.074	0.088			
-15.25	-0.363			0.074	0.071	0.077			

RUN.SEQ
182.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.818	1.994	4.54	1018	655	542.8	307.2	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.449	0.072	0.521	-0.0273	-0.0106	-0.0379	0.2254	31.08	39.63	32.27	43.23	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.349	0.081	0.430	-0.0453	-0.0132	-0.0585	37.97	41.28	38.59	0.617	0.139
0.296	0.441	0.082	0.523	-0.0459	-0.0208	-0.0667	35.40	50.44	37.75	0.638	-0.016
0.500	0.522	0.080	0.602	-0.0371	-0.0233	-0.0605	32.11	54.19	35.04	0.602	-0.111
0.697	0.553	0.073	0.626	-0.0328	-0.0227	-0.0555	30.93	56.18	33.86	0.493	-0.139
0.894	0.556	0.016	0.571	-0.0476	-0.0140	-0.0616	33.56	113.8	35.78	0.328	-0.106

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.543	0.543	0.543	0.543	0.543
0.01			0.397		
0.02			0.320		
0.03			0.270		
0.04			0.237		
0.05	0.220	0.209	0.207	0.187	0.065
0.10	0.150	0.129	0.072	0.104	0.011
0.15	0.109	0.093	0.059	-0.005	-0.035
0.20	0.086	0.011	0.031	0.020	-0.057
0.30	0.050	0.011	-0.010	-0.015	-0.092
0.40	-0.018	-0.006	-0.013	-0.026	-0.084
0.50	-0.014	-0.015	-0.013	-0.027	-0.120
0.55					
0.60	0.022	0.043	0.050	0.043	0.000
0.65			0.119		
0.70	0.110	0.151	0.157	0.098	0.099
0.75	0.133	0.135	0.176	0.159	0.137
0.80	0.147	0.177	0.186	0.175	0.156
0.85	0.108	0.170	0.143	0.176	0.163
0.90	0.115	0.137	0.147	0.154	0.098
0.95	0.062	0.084	0.100	0.132	0.076
1.00	-0.029	0.024	-0.001	0.026	-0.011

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.440	0.359	0.271	0.218	0.201
0.003			-0.182		
0.006			-0.368	-0.394	
0.01	-0.274	-0.508	-0.625	-0.647	-0.559
0.02	-0.576	-0.865	-0.953	-0.987	-0.787
0.03	-0.655	-0.029	-1.115	-1.070	-0.865
0.04	-0.640	-1.109	-1.167	-1.105	-0.847
0.05	-0.705	-1.144	-1.182	-1.246	-0.994
0.06	-0.600	-1.111	-1.279	-1.217	-1.060
0.08	-0.578	-1.064	-1.266	-1.204	-1.006
0.10	-0.553	-0.859	-1.270	-1.280	-1.019
0.125	-0.526	-0.887	-1.242	-1.232	-1.016
0.15	-0.508	-0.762	-1.110	-1.244	-1.010
0.175	-0.553	-0.710	-0.845	-1.179	-1.001
0.20	-0.485	-0.647	-0.726	-1.128	-1.070
0.225	-0.485	-0.597	-0.688	-0.926	-1.024
0.25	-0.455	-0.573	-0.665	-0.809	-1.025
0.30	-0.437	-0.620	-0.619	-0.667	-0.784
0.35	-0.411	-0.502	-0.601	-0.648	-0.776
0.40	-0.453	-0.478	-0.613	-0.613	-0.733
0.45	-0.423	-0.423	-0.526	-0.602	-0.691
0.50	-0.409	-0.406	-0.481	-0.547	-0.511
0.55	-0.376	-0.369	-0.418	-0.521	-0.412
0.60	-0.315	-0.387	-0.387	-0.380	-0.371
0.65	-0.269	-0.318	-0.326	-0.327	-0.330
0.70	-0.301	-0.270	-0.584	-0.275	-0.271
0.75	-0.253	-0.226	-0.234	-0.206	-0.293
0.80	-0.221	-0.182	-0.190	-0.158	-0.209
0.85	-0.165	-0.126	-0.129	-0.155	-0.175
0.90	-0.093	-0.419	-0.064	-0.076	-0.132
0.95	-0.036	-0.027	0.008	-0.020	-0.074
1.00	-0.029	0.024	-0.001	0.026	-0.011

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	-0.031				-0.389				
11.35	0.270	0.100				-0.386				
12.35	0.294					-0.395		-0.343		
14.35	0.342	0.071				-0.395				
15.35	0.366	0.065				-0.396				-0.336
16.35	0.390	0.054				-0.397				1.968
17.35	0.413	0.051				-0.385		-0.371		-0.310
18.35	0.437	0.042				-0.383				-0.323
19.35	0.461	0.037				-0.378				-0.315
20.35	0.485	0.045				-0.376				-0.320
22.35	0.533	0.046				-0.349				-0.305
23.35	0.556	0.047				-0.360				-0.297
24.35	0.580	0.053				-0.320		-0.320		-0.304
25.35	0.604	0.056				-0.308				-0.290
26.35	0.628	0.066				-0.302				-0.279
27.35	0.652	0.081				-0.295				
30.35	0.723									-0.223
31.35	0.747		0.140		-0.224		-0.233	-0.239	-0.227	-0.213
32.35	0.771		0.143		-0.204		-0.224		-0.221	-0.202
33.35	0.795		0.148		-0.208		-0.204		-0.193	-0.183
34.35	0.818		0.147		-0.176		-0.177		-0.185	-0.185
35.35	0.842		0.144		-0.138		-0.172		-0.170	-0.175
36.35	0.866		0.148		-0.103		-0.157		-0.147	-0.158
37.35	0.890		0.129		-0.093		-0.116		-0.123	-0.125
38.35	0.914		0.105		-0.069		-0.094		-0.094	-0.116
39.35	0.938		0.105	-0.047	-0.045		-0.078		-0.074	-0.121
40.35	0.961		0.076	-0.020	-0.024		-0.065		-0.094	-0.108
41.35	0.985		0.040	0.016	0.008		-0.047		-0.075	-0.103
42.35	1.009		0.071	0.023	0.002		-0.028		-0.061	-0.175
44.85	1.069		0.052	0.020	0.008		-0.020		-0.039	-0.076
45.85	1.092		0.041	0.026	0.012					
46.85	1.116		0.002	0.028	0.011					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.400	-0.203	-0.162			
13.75	0.328		-0.303	-0.279	-0.208	-0.153			
10.75	0.256		-0.310	-0.304	-0.213	-0.158	-0.103	-0.175	
7.75	0.185	-0.343	-0.371	-0.320	-0.239	-0.156	-0.092		-0.055
6.75	0.161				-0.227	-0.147	-0.075	-0.061	-0.039
5.75	0.137	-0.387	-0.391	-0.322	-0.253	-0.151	-0.079	-0.044	-0.030
4.75	0.113				-0.258	-0.137	-0.047	-0.040	-0.014
4.25	0.101	-0.395	-0.385	-0.320					
3.75	0.089				-0.233	-0.157	-0.047	-0.028	-0.020
2.75	0.066				-0.252	-0.149	-0.028	-0.011	-0.000
1.75	0.042				-0.241	0.037	-0.014	-0.015	0.003
0.75	0.018				-0.224	-0.103	0.008	0.002	0.008
-0.25	-0.006						0.016	0.023	0.020
-1.25	-0.030					0.140	0.148	0.040	0.071	0.052
-2.25	-0.054					0.131	0.120	0.063	0.059	0.031
-2.75	-0.066			0.051	0.053					
-3.25	-0.077						0.125	0.106	0.067	0.061	0.026
-4.25	-0.101	0.067	0.048	0.053	0.098	0.088	0.071	0.068	0.042
-5.25	-0.125					0.085	0.098	0.081	0.060	0.043
-6.25	-0.149	0.081	0.063	0.054	0.100	0.102	0.069		-0.131
-9.25	-0.220		0.079	0.058	0.071	0.093	0.074		
-12.25	-0.292		0.070	0.054	0.064	0.071			
-15.25	-0.363			0.055	0.077	0.056			

RUN.SEQ
183.1

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.820	1.494	3.40	745	479	533.4	225.3	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.435	0.077	0.512	-0.0212	-0.0119	-0.0331	0.2205	29.88	40.30	31.46	43.06	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.340	0.092	0.432	-0.0419	-0.0175	-0.0595	37.33	44.05	38.76	0.619	0.138
0.296	0.424	0.087	0.512	-0.0310	-0.0225	-0.0536	32.31	50.84	35.47	0.624	0.002
0.500	0.493	0.083	0.576	-0.0293	-0.0245	-0.0539	30.95	54.50	34.35	0.576	-0.103
0.697	0.544	0.070	0.614	-0.0319	-0.0244	-0.0562	30.86	59.83	34.17	0.483	-0.137
0.894	0.551	0.020	0.571	-0.0470	-0.0161	-0.0631	33.53	106.6	36.04	0.328	-0.107

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.543	0.543	0.543	0.543	0.543
0.01			0.388		
0.02			0.315		
0.03			0.277		
0.04			0.238		
0.05	0.215	0.219	0.197	0.171	0.078
0.10	0.134	0.121	0.094	0.083	0.002
0.15	0.104	0.084	0.054	0.032	-0.043
0.20	0.104	0.066	0.034	-0.004	-0.052
0.30	0.057	-0.008	0.003	-0.039	-0.104
0.40	0.024	-0.024	-0.011	-0.044	-0.101
0.50	-0.011	-0.021	-0.005	-0.026	-0.087
0.55					
0.60	0.020	0.050	0.039	0.046	0.002
0.65			0.093		
0.70	0.112	0.151	0.133	0.152	0.111
0.75	0.145	0.169	0.187	0.160	0.152
0.80	0.160	0.179	0.193	0.177	0.153
0.85	0.151	0.171	0.186	0.188	0.155
0.90	0.128	0.140	0.145	0.164	0.127
0.95	0.076	0.108	0.096	0.122	0.085
1.00	0.012	0.038	0.028	0.012	-0.020

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.440	0.357	0.275	0.221	0.244
0.003			-0.135		
0.006			-0.360	-0.383	
0.01	-0.282	-0.502	-0.623	-0.613	-0.554
0.02	-0.563	-0.860	-0.935	-0.913	-0.779
0.03	-0.641	-0.952	-1.087	-1.063	-0.844
0.04	-0.636	-1.110	-1.149	-1.106	-0.846
0.05	-0.654	-1.146	-1.241	-1.210	-0.976
0.06	-0.601	-1.103	-1.261	-1.205	-0.981
0.08	-0.576	-1.041	-1.256	-1.199	-0.992
0.10	-0.542	-0.965	-1.247	-1.210	-1.006
0.125	-0.525	-0.759	-1.211	-1.210	-0.986
0.15	-0.519	-0.748	-1.128	-1.172	-0.971
0.175	-0.508	-0.700	-0.797	-1.122	-0.982
0.20	-0.500	-0.624	-0.702	-1.114	-0.990
0.225	-0.492	-0.589	-0.675	-1.064	-1.028
0.25	-0.457	-0.562	-0.632	-0.796	-1.025
0.30	-0.448	-0.527	-0.595	-0.660	-0.955
0.35	-0.420	-0.524	-0.553	-0.668	-0.893
0.40	-0.417	-0.490	-0.554	-0.632	-0.693
0.45	-0.407	-0.442	-0.524	-0.589	-0.551
0.50	-0.389	-0.411	-0.477	-0.540	-0.493
0.55	-0.359	-0.362	-0.410	-0.462	-0.433
0.60	-0.321	-0.342	-0.380	-0.381	-0.372
0.65	-0.272	-0.325	-0.323	-0.328	-0.303
0.70	-0.259	-0.279	-0.273	-0.262	-0.258
0.75	-0.245	-0.224	-0.234	-0.216	-0.240
0.80	-0.212	-0.175	-0.195	-0.171	-0.228
0.85	-0.144	-0.124	-0.122	-0.117	-0.184
0.90	-0.094	-0.064	-0.061	-0.070	-0.129
0.95	-0.031	-0.008	-0.005	-0.017	-0.090
1.00	0.012	0.038	0.028	0.012	-0.020

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	-0.080				-0.391				
11.35	0.270	0.093				-0.406				
12.35	0.294					-0.387		-0.357		
14.35	0.342	0.082				-0.393				
15.35	0.366	0.069				-0.397				-0.314
16.35	0.390	0.052				-0.390				2.154
17.35	0.413	0.045				-0.364		-0.334		-0.294
18.35	0.437	0.052				-0.354				-0.311
19.35	0.461	0.046				-0.369				-0.296
20.35	0.485	0.032				-0.372				-0.277
22.35	0.533	0.020				-0.367				-0.306
23.35	0.556	0.021				-0.333				-0.265
24.35	0.580	0.035				-0.310		-0.320		-0.281
25.35	0.604	0.045				-0.321				-0.286
26.35	0.628	0.057				-0.301				-0.281
27.35	0.652	0.085				-0.289				
30.35	0.723									-0.229
31.35	0.747		0.138		-0.245		-0.239	-0.229	-0.215	-0.179
32.35	0.771		0.137		-0.226		-0.215		-0.185	-0.174
33.35	0.795		0.138		-0.201		-0 199		-0.191	-0.195
34.35	0.818		0.146		-0.166		-0.178		-0.160	-0.192
35.35	0.842		0.134		-0.150		-0.160		-0.163	-0.160
36.35	0.866		0.134		-0.114		-0.129		-0.151	-0.141
37.35	0.890		0.116		-0.093		-0.110		-0.132	-0.114
38.35	0.914		0.089		-0.070		-0.107		-0.106	-0.129
39.35	0.938		0.102	-0.045	-0.057		-0.079		-0.105	-0.121
40.35	0.961		0.080	-0.009	-0.019		-0.051		-0.066	-0.108
41.35	0.985		0.053	0.023	-0.003		-0.020		-0.067	-0.090
42.35	1.009		0.043	0.012	0.024		-0.014		-0.059	-0.206
44.85	1.069		0.026	0.007	0.014		-0.018		-0.045	-0.066
45.85	1.092		0.025	0.015	-0.005					
46.85	1.116		0.025	0.021	-0.001					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.444	-0.184	-0.114			
13.75	0.328		-0.259	-0.270	-0.212	-0.123			
10.75	0.256		-0.294	-0.281	-0.179	-0.141	-0.090	-0.206	
7.75	0.185	-0.357	-0.334	-0.320	-0.229	-0.138	-0.063		-0.041
6.75	0.161					-0.215	-0.151	-0.067	-0.059 -0.045
5.75	0.137	-0.379	-0.374	-0.338	-0.224	-0.141	-0.065	-0.031	-0.042
4.75	0.113					-0.228	-0.125	-0.057	-0.041 -0.026
4.25	0.101	-0.387	-0.364	-0.310					
3.75	0.089					-0.239	-0.129	-0.020	-0.014 -0.018
2.75	0.066					-0.248	-0.127	-0.036	-0.017 0.001
1.75	0.042					-0.210	-0.129	-0.026	0.003 -0.003
0.75	0.018					-0.245	-0.114	-0.003	0.024 0.014
-0.25	-0.006							0.023	0.012 0.007
-1.25	-0.030						0.138	0.134	0.053 0.043 0.026
-2.25	-0.054						0.111	0.108	0.052 0.044 0.038
-2.75	-0.066			0.045	0.035				
-3.25	-0.077							0.093	0.101 0.064 0.067 0.034
-4.25	-0.101	0.090	0.052	0.053	0.084	0.083	0.074	0.067	0.031
-5.25	-0.125						0.080	0.096	0.067 0.059 0.052
-6.25	-0.149	0.080	0.060	0.043	0.094	0.088	0.065		-0.221
-9.25	-0.220		0.059	0.062	0.081	0.098	0.064		
-12.25	-0.292		0.066	0.060	0.059	0.079			
-15.25	-0.363			0.054	0.061	0.057			

RUN.SEQ
184.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.857	1.987	4.52	977	605	536.4	311.0	5.00	17

WING COEFFICIENTS

	CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
	0.456	0.083	0.539	-0.0283	-0.0126	-0.0409	0.2329	31.21	40.16	32.59	43.17	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.355	0.101	0.456	-0.0505	-0.0190	-0.0696	39.25	43.83	40.27	0.653	0.131
0.296	0.442	0.095	0.537	-0.0386	-0.0252	-0.0638	33.74	51.40	36.88	0.655	-0.009
0.500	0.509	0.091	0.600	-0.0320	-0.0284	-0.0604	31.28	56.15	35.05	0.600	-0.111
0.697	0.574	0.074	0.648	-0.0336	-0.0258	-0.0593	30.84	59.94	34.16	0.510	-0.145
0.894	0.613	0.005	0.618	-0.0661	-0.0135	-0.0796	35.79	293.8	37.88	0.355	-0.119

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.548	0.548	0.548	0.548	0.548
0.01			0.386		
0.02			0.308		
0.03			0.261		
0.04			0.229		
0.05	0.238	0.218	0.196	0.164	0.049
0.10	0.167	0.132	0.105	0.083	-0.007
0.15	0.128	0.086	0.064	0.023	-0.051
0.20	0.097	0.062	0.044	0.009	-0.080
0.30	0.060	0.010	0.003	-0.026	-0.121
0.40	0.033	-0.005	-0.014	-0.038	-0.115
0.50	0.001	-0.016	-0.010	-0.038	-0.098
0.55					
0.60	0.029	0.048	0.042	0.051	-0.022
0.65			0.102		
0.70	0.119	0.156	0.153	0.149	0.093
0.75	0.161	0.189	0.196	0.175	0.129
0.80	0.169	0.201	0.213	0.190	0.157
0.85	0.160	0.193	0.210	0.189	0.161
0.90	0.125	0.152	0.169	0.161	0.133
0.95	0.077	0.103	0.121	0.127	0.076
1.00	0.027	0.046	0.051	0.048	-0.066

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.464	0.408	0.330	0.284	0.297
0.003			-0.053		
0.006			-0.264	-0.288	
0.01	-0.207	-0.407	-0.506	-0.518	-0.445
0.02	-0.498	-0.752	-0.819	-0.793	-0.663
0.03	-0.593	-0.892	-0.971	-0.932	-0.740
0.04	-0.589	-0.996	-1.029	-0.965	-0.718
0.05	-0.601	-1.043	-1.114	-1.090	-0.876
0.06	-0.565	-1.032	-1.139	-1.084	-0.885
0.08	-0.543	-1.025	-1.136	-1.079	-0.891
0.10	-0.515	-0.934	-1.145	-1.097	-0.910
0.125	-0.493	-0.757	-1.135	-1.116	-0.916
0.15	-0.477	-0.755	-1.120	-1.137	-0.933
0.175	-0.473	-0.703	-1.100	-1.129	-0.945
0.20	-0.486	-0.646	-0.844	-1.129	-0.971
0.225	-0.478	-0.610	-0.735	-1.124	-0.990
0.25	-0.447	-0.586	-0.689	-1.107	-1.007
0.30	-0.450	-0.557	-0.635	-1.061	-1.060
0.35	-0.440	-0.527	-0.608	-0.697	-1.134
0.40	-0.443	-0.518	-0.608	-0.715	-1.179
0.45	-0.438	-0.496	-0.600	-0.735	-0.687
0.50	-0.439	-0.491	-0.573	-0.715	-0.640
0.55	-0.436	-0.464	-0.525	-0.625	-0.602
0.60	-0.390	-0.419	-0.435	-0.328	-0.521
0.65	-0.375	-0.369	-0.306	-0.247	-0.437
0.70	-0.339	-0.292	-0.249	-0.205	-0.306
0.75	-0.262	-0.233	-0.203	-0.170	-0.231
0.80	-0.214	-0.187	-0.161	-0.131	-0.213
0.85	-0.157	-0.119	-0.102	-0.079	-0.163
0.90	-0.085	-0.071	-0.044	-0.051	-0.123
0.95	-0.017	-0.002	0.013	0.004	-0.092
1.00	0.027	0.046	0.051	0.048	-0.066

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	-0.051				-0.394				
11.35	0.270	0.069				-0.408				
12.35	0.294					-0.409		-0.354		
14.35	0.342	0.056				-0.410				
15.35	0.366	0.053				-0.413				-0.354
16.35	0.390	0.051				-0.424				0.949
17.35	0.413	0.030				-0.411		-0.383		-0.342
18.35	0.437	0.029				-0.422				-0.345
19.35	0.461	0.013				-0.408				-0.355
20.35	0.485	0.020				-0.414				-0.348
22.35	0.533	0.015				-0.420				-0.359
23.35	0.556	0.031				-0.418				-0.333
24.35	0.580	0.039				-0.372		-0.404		-0.334
25.35	0.604	0.038				-0.380				-0.356
26.35	0.628	0.043				-0.365				-0.345
27.35	0.652	0.076				-0.342				
30.35	0.723									-0.262
31.35	0.747		0.134		-0.272		-0.281	-0.230	-0.290	-0.302
32.35	0.771		0.138		-0.268		-0.260		-0.256	-0.263
33.35	0.795		0.125		-0.222		-0.229		-0.242	-0.237
34.35	0.818		0.134		-0.197		-0.195		-0.197	-0.216
35.35	0.842		0.120		-0.193		-0.189		-0.176	-0.179
36.35	0.866		0.116		-0.130		-0.183		-0.158	-0.175
37.35	0.890		0.104		-0.118		-0.152		-0.156	-0.162
38.35	0.914		0.089		-0.096		-0.121		-0.125	-0.138
39.35	0.938		0.075	-0.044	-0.066		-0.093		-0.105	-0.116
40.35	0.961		0.071	-0.016	-0.031		-0.066		-0.075	-0.136
41.35	0.985		0.028	0.008	-0.013		-0.071		-0.043	-0.119
42.35	1.009		0.049	0.008	0.001		-0.036		-0.013	-0.200
44.85	1.069		0.029	0.006	0.005		-0.024		-0.063	-0.087
45.85	1.092		0.031	0.004	-0.004					
46.85	1.116		0.010	0.014	-0.019					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.466	-0.240	-0.200			
13.75	0.328		-0.318	-0.343	-0.237	-0.174			
10.75	0.256		-0.342	-0.334	-0.302	-0.175	-0.119	-0.200	
7.75	0.185	-0.354	-0.383	-0.404	-0.230	-0.159	-0.079		-0.056
6.75	0.161				-0.290	-0.158	-0.043	-0.013	-0.063
5.75	0.137	-0.392	-0.415	-0.425	-0.249	-0.158	-0.071	-0.039	-0.045
4.75	0.113				-0.284	-0.154	-0.071	-0.034	-0.053
4.25	0.101	-0.409	-0.411	-0.372					
3.75	0.089				-0.281	-0.183	-0.071	-0.036	-0.024
2.75	0.066				-0.304	-0.145	-0.038	-0.010	-0.035
1.75	0.042				-0.304	-0.175	-0.028	-0.013	-0.013
0.75	0.018				-0.272	-0.130	-0.013	0.001	0.005
-0.25	-0.006						0.008	0.008	0.006
-1.25	-0.030					0.134	0.116	0.028	0.049	0.029
-2.25	-0.054					0.109	0.112	0.062	0.047	0.042
-2.75	-0.066			0.030	0.039					
-3.25	-0.077					0.104	0.107	0.056	0.067	0.031
-4.25	-0.101	0.070	0.035	0.042	0.103	0.081	0.047	0.065	0.021
-5.25	-0.125				0.089	0.072	0.059	0.059	0.037
-6.25	-0.149	0.075	0.043	0.047	0.080	0.077	0.070		-0.152
-9.25	-0.220		0.061	0.054	0.063	0.065	0.082		
-12.25	-0.292		0.054	0.066	0.049	0.053			
-15.25	-0.363			0.055	0.059	0.053			

RUN.SEQ
185.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.962	2.973	6.76	1426	787	544.1	509.9	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.528	0.043	0.570	-0.0969	0.0000	-0.0969	0.2448	43.37	24.97	42.00	42.93	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.398	0.083	0.482	-0.1009	-0.0116	-0.1125	50.33	38.89	48.35	0.690	0.059
0.296	0.500	0.061	0.570	-0.1071	-0.0138	-0.1209	46.03	47.67	46.21	0.696	-0.089
0.500	0.599	0.043	0.642	-0.1125	-0.0161	-0.1286	43.79	62.45	45.05	0.641	-0.183
0.697	0.693	0.012	0.705	-0.1409	-0.0090	-0.1499	45.33	97.41	46.25	0.555	-0.211
0.894	0.673	-0.083	0.590	-0.1646	0.0102	-0.1544	49.46	37.27	51.18	0.338	-0.139

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.561	0.561	0.561	0.561	0.561
0.01			0.330		
0.02			0.267		
0.03			0.225		
0.04			0.192		
0.05	0.243	0.194	0.158	0.104	-0.035
0.10	0.163	0.118	0.066	0.024	-0.089
0.15	0.123	0.072	0.023	-0.028	-0.122
0.20	0.099	0.041	-0.002	-0.055	-0.134
0.30	0.053	-0.011	-0.053	-0.093	-0.200
0.40	0.007	-0.043	-0.080	-0.105	-0.267
0.50	-0.035	-0.082	-0.084	-0.102	-0.279
0.55					
0.60	-0.003	-0.006	0.013	-0.020	-0.072
0.65			0.077		
0.70	0.101	0.127	0.129	0.103	0.056
0.75	0.142	0.165	0.171	0.142	0.094
0.80	0.161	0.182	0.183	0.162	0.111
0.85	0.153	0.175	0.175	0.158	0.109
0.90	0.112	0.123	0.128	0.122	0.073
0.95	0.037	0.049	0.049	0.038	-0.015
1.00	-0.067	-0.078	-0.070	-0.222	-0.488

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.575	0.513	0.458	0.432	0.444
0.003			0.144		
0.006			-0.035	-0.038	
0.01	-0.055	-0.185	-0.240	-0.231	-0.125
0.02	-0.339	-0.495	-0.525	-0.480	-0.314
0.03	-0.460	-0.649	-0.667	-0.604	-0.390
0.04	-0.452	-0.725	-0.718	-0.625	-0.410
0.05	-0.491	-0.778	-0.790	-0.769	-0.605
0.06	-0.463	-0.776	-0.818	-0.772	-0.602
0.08	-0.436	-0.792	-0.830	-0.751	-0.555
0.10	-0.419	-0.810	-0.861	-0.785	-0.586
0.125	-0.405	-0.705	-0.862	-0.816	-0.618
0.15	-0.390	-0.639	-0.882	-0.844	-0.639
0.175	-0.384	-0.598	-0.885	-0.849	-0.662
0.20	-0.395	-0.585	-0.875	-0.858	-0.693
0.225	-0.388	-0.553	-0.862	-0.864	-0.728
0.25	-0.374	-0.524	-0.805	-0.864	-0.745
0.30	-0.395	-0.504	-0.636	-0.888	-0.809
0.35	-0.383	-0.486	-0.590	-0.902	-0.865
0.40	-0.395	-0.489	-0.587	-0.912	-0.933
0.45	-0.409	-0.485	-0.586	-0.887	-0.983
0.50	-0.414	-0.483	-0.581	-0.717	-0.864
0.55	-0.425	-0.486	-0.578	-0.691	-0.690
0.60	-0.425	-0.490	-0.579	-0.690	-0.610
0.65	-0.400	-0.490	-0.574	-0.692	-0.614
0.70	-0.455	-0.484	-0.572	-0.647	-0.610
0.75	-0.468	-0.499	-0.557	-0.602	-0.615
0.80	-0.482	-0.522	-0.574	-0.575	-0.606
0.85	-0.487	-0.526	-0.574	-0.557	-0.600
0.90	-0.459	-0.526	-0.458	-0.493	-0.590
0.95	-0.245	-0.278	-0.173	-0.316	-0.576
1.00	-0.067	-0.078	-0.070	-0.222	-0.488

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.039	.185	.161	.256
X	X/CR									
10.35	0.247	0.014				-0.293				
11.35	0.270	0.088				-0.308				
12.35	0.294					-0.322		-0.270		
14.35	0.342	0.066				-0.333				
15.35	0.366	0.054				-0.340				-0.284
16.35	0.390	0.041				-0.347				0.132
17.35	0.413	0.033				-0.349		-0.318		-0.280
18.35	0.437	0.015				-0.360				-0.290
19.35	0.461	0.011				-0.370				-0.302
20.35	0.485	0.009				-0.374				-0.305
22.35	0.533	-0.003				-0.379				-0.328
23.35	0.556	-0.001				-0.399				-0.325
24.35	0.580	0.007				-0.400		-0.368		-0.340
25.35	0.604	0.007				-0.407				-0.349
26.35	0.628	0.024				-0.393				-0.355
27.35	0.652	0.050				-0.392				
30.35	0.723									-0.365
31.35	0.747		0.123		-0.457		-0.433	-0.393	-0.406	-0.365
32.35	0.771		0.133		-0.459		-0.435		-0.413	-0.368
33.35	0.795		0.134		-0.469		-0.[illegible]		-0.421	-0.377
34.35	0.818		0.137		-0.472		-0.[illegible]		-0.418	-0.381
35.35	0.842		0.137		-0.474		-0.454		-0.423	-0.382
36.35	0.866		0.119		-0.465		-0.464		-0.433	-0.391
37.35	0.890		0.096		-0.433		-0.462		-0.438	-0.392
38.35	0.914		0.062		-0.392		-0.440		-0.437	-0.398
39.35	0.938		0.043	-0.273	-0.300		-0.379		-0.426	-0.394
40.35	0.961		0.004	-0.182	-0.219		-0.302		-0.398	-0.408
41.35	0.985		-0.066	-0.105	-0.152		-0.223		-0.316	-0.388
42.35	1.009		-0.027	-0.078	-0.091		-0.170		-0.229	-0.397
44.85	1.069		-0.021	-0.030	-0.037		-0.083		-0.092	-0.292
45.85	1.092		-0.017	-0.018	-0.023					
46.85	1.116		-0.010	-0.036	-0.008					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.352	-0.321	-0.336			
13.75	0.328		-0.245	-0.309	-0.345	-0.357			
10.75	0.256		-0.280	-0.340	-0.365	-0.391	-0.388	-0.397	
7.75	0.185	-0.270	-0.318	-0.368	-0.393	-0.417	-0.340		-0.092
6.75	0.161					-0.406	-0.433	-0.316	-0.229 -0.092
5.75	0.137	-0.303	-0.342	-0.393	-0.413	-0.433	-0.242	-0.208	-0.060
4.75	0.113					-0.417	-0.436	-0.238	-0.172 -0.051
4.25	0.101	-0.322	-0.349	-0.400					
3.75	0.089					-0.433	-0.464	-0.223	-0.170 -0.083
2.75	0.066					-0.439	-0.451	-0.195	-0.126 -0.026
1.75	0.042					-0.448	-0.471	-0.172	-0.111 -0.018
0.75	0.018					-0.457	-0.465	-0.152	-0.091 -0.037
-0.25	-0.006							-0.105	-0.078 -0.030
-1.25	-0.030						0.123	0.119	-0.066 -0.027 -0.021
-2.25	-0.054						0.105	0.109	-0.001 0.037 0.000
-2.75	-0.066			0.033	0.007				
-3.25	-0.077						0.090	0.098	0.017 0.037 0.007
-4.25	-0.101	0.078	0.031	0.004	0.081	0.079	0.023	0.035	0.009
-5.25	-0.125					0.083	0.064	0.034	0.024 0.012
-6.25	-0.149	0.085	0.044	0.014	0.070	0.061	0.027		-0.095
-9.25	-0.220		0.057	0.027	0.055	0.064	0.039		
-12.25	-0.292		0.060	0.037	0.052	0.062			
-15.25	-0.363			0.040	0.041	0.050			

RUN.SEQ
186.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.948	2.983	6.79	1443	810	545.5	508.9	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.514	0.054	0.568	-0.0860	-0.0033	-0.0894	0.2425	41.74	31.20	40.74	42.71	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.391	0.088	0.480	-0.0933	-0.0140	-0.1073	48.83	40.79	47.35	0.688	0.068
0.296	0.497	0.074	0.571	-0.0946	-0.0186	-0.1132	44.06	50.06	44.83	0.696	-0.077
0.500	0.587	0.060	0.646	-0.1018	-0.0210	-0.1227	42.35	60.09	43.99	0.646	-0.177
0.697	0.687	0.023	0.709	-0.1319	-0.0108	-0.1426	44.21	71.85	45.10	0.5[illegible]	-0.207
0.894	0.612	-0.067	0.545	-0.1372	0.0079	-0.1292	47.42	36.87	48.71	0.313	-0.124

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.559	0.559	0.559	0.559	0.559
0.01			0.342		
0.02			0.281		
0.03			0.238		
0.04			0.203		
0.05	0.244	0.204	0.175	0.115	-0.014
0.10	0.167	0.123	0.084	0.033	-0.069
0.15	0.128	0.080	0.033	-0.009	-0.106
0.20	0.099	0.050	0.009	-0.040	-0.121
0.30	0.052	-0.003	-0.039	-0.076	-0.192
0.40	0.014	-0.031	-0.061	-0.085	-0.237
0.50	-0.032	-0.064	-0.065	-0.096	-0.188
0.55					
0.60	-0.000	0.007	0.034	-0.005	-0.069
0.65			0.093		
0.70	0.112	0.142	0.142	0.114	0.055
0.75	0.148	0.180	0.178	0.148	0.092
0.80	0.164	0.187	0.195	0.171	0.109
0.85	0.159	0.185	0.194	0.169	0.102
0.90	0.124	0.140	0.144	0.123	0.067
0.95	0.051	0.071	0.077	0.039	-0.028
1.00	-0.032	0.001	-0.024	-0.240	-0.421

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.578	0.500	0.436	0.408	0.421
0.003			0.117		
0.006			-0.064	-0.072	
0.01	-0.054	-0.213	-0.273	-0.270	-0.162
0.02	-0.347	-0.526	-0.558	-0.519	-0.352
0.03	-0.474	-0.671	-0.701	-0.639	-0.424
0.04	-0.464	-0.756	-0.755	-0.663	-0.438
0.05	-0.496	-0.807	-0.824	-0.797	-0.640
0.06	-0.473	-0.800	-0.850	-0.804	-0.637
0.08	-0.448	-0.810	-0.861	-0.785	-0.596
0.10	-0.431	-0.830	-0.886	-0.813	-0.625
0.125	-0.413	-0.692	-0.893	-0.843	-0.650
0.15	-0.399	-0.648	-0.906	-0.871	-0.666
0.175	-0.394	-0.611	-0.907	-0.873	-0.695
0.20	-0.404	-0.594	-0.890	-0.883	-0.720
0.225	-0.397	-0.560	-0.871	-0.889	-0.756
0.25	-0.381	-0.532	-0.741	-0.887	-0.772
0.30	-0.400	-0.509	-0.626	-0.911	-0.835
0.35	-0.390	-0.495	-0.591	-0.922	-0.903
0.40	-0.402	-0.497	-0.589	-0.928	-0.892
0.45	-0.417	-0.491	-0.591	-0.844	-0.613
0.50	-0.423	-0.487	-0.585	-0.713	-0.567
0.55	-0.429	-0.492	-0.579	-0.698	-0.558
0.60	-0.433	-0.492	-0.579	-0.693	-0.548
0.65	-0.405	-0.485	-0.573	-0.657	-0.539
0.70	-0.461	-0.477	-0.566	-0.570	-0.549
0.75	-0.460	-0.498	-0.552	-0.548	-0.546
0.80	-0.471	-0.510	-0.570	-0.536	-0.518
0.85	-0.470	-0.520	-0.560	-0.500	-0.519
0.90	-0.362	-0.421	-0.300	-0.424	-0.515
0.95	-0.156	-0.099	-0.091	-0.329	-0.500
1.00	-0.032	0.001	-0.024	-0.240	-0.421

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.027				-0.306				
11.35	0.270	0.098				-0.320				
12.35	0.294					-0.336		-0.283		
14.35	0.342	0.086				-0.344				
15.35	0.366	0.069				0.350				-0.301
16.35	0.390	0.057				-0.366				0.056
17.35	0.413	0.042				-0.365		-0.330		-0.293
18.35	0.437	0.027				-0.371				-0.304
19.35	0.461	0.027				-0.378				-0.310
20.35	0.485	0.021				-0.384				-0.315
22.35	0.533	0.017				-0.390				-0.335
23.35	0.556	0.013				-0.406				-0.333
24.35	0.580	0.020				-0.403		-0.379		-0.348
25.35	0.604	0.031				-0.411				-0.354
26.35	0.628	0.043				-0.404				-0.365
27.35	0.652	0.052				-0.404				
30.35	0.723									-0.376
31.35	0.747		0.127		-0.457		-0.443	-0.406	-0.411	-0.367
32.35	0.771		0.144		-0.457		-0.449		-0.414	-0.369
33.35	0.795		0.143		-0.468		-0.455		-0.419	-0.373
34.35	0.818		0.139		-0.470		-0.449		-0.425	-0.385
35.35	0.842		0.129		-0.471		-0.449		-0.433	-0.387
36.35	0.866		0.130		-0.447		-0.448		-0.439	-0.397
37.35	0.890		0.112		-0.386		-0.434		-0.430	-0.404
38.35	0.914		0.087		-0.310		-0.388		-0.391	-0.414
39.35	0.938		0.066	-0.194	-0.176		-0.330		-0.364	-0.411
40.35	0.961		0.040	-0.126	-0.141		-0.252		-0.296	-0.368
41.35	0.985		0.002	-0.064	-0.092		-0.079		-0.199	-0.277
42.35	1.009		0.031	-0.042	-0.[illegible]		-0.035		-0.115	-0.236
44.85	1.069		0.028	0.009	-0.[illegible]		-0.009		-0.077	-0.159
45.85	1.092		0.024	0.008	-0.007					
46.85	1.116		0.029	-0.008	0.006					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.361	-0.327	-0.352			
13.75	0.328		-0.256	-0.318	-0.353	-0.372			
10.75	0.256		-0.293	-0.348	-0.367	-0.397	-0.277	-0.236	
7.75	0.185	-0.283	-0.330	-0.379	-0.406	-0.432	-0.274		-0.083
6.75	0.161				-0.411	-0.439	-0.199	-0.115	-0.077
5.75	0.137	-0.315	-0.359	-0.402	-0.430	-0.447	-0.225	-0.147	-0.066
4.75	0.113				-0.434	-0.458	-0.138	-0.128	-0.067
4.25	0.101	-0.336	-0.365	-0.403					
3.75	0.089				-0.443	-0.448	-0.079	-0.035	-0.009
2.75	0.066				-0.455	-0.471	-0.144	-0.095	-0.041
1.75	0.042				-0.460	-0.456	-0.121	-0.087	-0.018
0.75	0.018				-0.457	-0.447	-0.092	-0.059	-0.024
-0.25	-0.006						-0.064	-0.042	0.009
-1.25	-0.030					0.127	0.130	0.002	0.031	0.028
-2.25	-0.054					0.114	0.113	0.022	0.037	0.018
-2.75	-0.066		0.042	0.020						
-3.25	-0.077					0.104	0.107	0.026	0.035	0.017
-4.25	-0.101	0.090	0.040	0.014	0.094	0.094	0.030	0.041	0.026
-5.25	-0.125				0.084	0.088	0.065	0.036	0.024
-6.25	-0.149	0.094	0.050	0.018	0.081	0.085	0.051		-0.082
-9.25	-0.220		0.066	0.031	0.062	0.062	0.051		
-12.25	-0.292		0.063	0.036	0.056	0.054			
-15.25	-0.363			0.040	0.048	0.044			

RUN.SEQ
187.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.940	3.000	6.82	1456	824	545.8	509.6	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.510	0.056	0.566	-0.0802	-0.0040	-0.0842	0.2411	40.73	32.09	39.87	42.58	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.387	0.091	0.479	-0.0885	-0.0150	-0.1035	47.84	41.44	46.62	0.686	0.075
0.296	0.503	0.075	0.578	-0.0939	-0.0192	-0.1131	43.69	50.61	44.59	0.705	-0.076
0.500	0.583	0.059	0.642	-0.0938	-0.0205	-0.1143	41.08	60.11	42.81	0.641	-0.169
0.697	0.671	0.031	0.701	-0.1194	-0.0119	-0.1313	42.81	63.70	43.73	0.552	-0.199
0.894	0.598	-0.062	0.536	-0.1291	0.0070	-0.1220	46.59	36.40	47.77	0.308	-0.121

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.558	0.558	0.558	0.558	0.558
0.01			0.332		
0.02			0.269		
0.03			0.229		
0.04			0.196		
0.05	0.245	0.196	0.168	0.120	-0.001
0.10	0.169	0.118	0.078	0.045	-0.065
0.15	0.130	0.080	0.033	-0.006	-0.100
0.20	0.098	0.052	0.011	-0.028	-0.118
0.30	0.052	-0.002	-0.035	-0.061	-0.197
0.40	0.015	-0.029	-0.062	-0.076	-0.229
0.50	-0.021	-0.057	-0.063	-0.084	-0.162
0.55					
0.60	0.011	0.015	0.031	0.004	-0.068
0.65			0.098		
0.70	0.107	0.135	0.147	0.120	0.056
0.75	0.147	0.167	0.176	0.152	0.094
0.80	0.163	0.194	0.190	0.167	0.101
0.85	0.162	0.188	0.183	0.162	0.099
0.90	0.120	0.137	0.137	0.127	0.063
0.95	0.057	0.077	0.068	0.038	-0.023
1.00	-0.009	0.008	-0.007	-0.202	-0.391

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.564	0.488	0.425	0.395	0.409
0.003			0.100		
0.006			-0.085	-0.094	
0.01	-0.078	-0.235	-0.296	-0.294	-0.194
0.02	-0.366	-0.549	-0.584	-0.544	-0.388
0.03	-0.486	-0.696	-0.728	-0.666	-0.458
0.04	-0.481	-0.776	-0.777	-0.691	-0.464
0.05	-0.501	-0.829	-0.848	-0.834	-0.662
0.06	-0.484	-0.823	-0.877	-0.829	-0.663
0.08	-0.457	-0.830	-0.884	-0.812	-0.621
0.10	-0.438	-0.850	-0.914	-0.838	-0.653
0.125	-0.423	-0.712	-0.917	-0.868	-0.674
0.15	-0.408	-0.654	-0.924	-0.896	-0.695
0.175	-0.405	-0.628	-0.931	-0.901	-0.720
0.20	-0.410	-0.608	-0.904	-0.908	-0.744
0.225	-0.402	-0.566	-0.881	-0.913	-0.777
0.25	-0.387	-0.538	-0.739	-0.910	-0.795
0.30	-0.401	-0.512	-0.635	-0.931	-0.860
0.35	-0.391	-0.500	-0.602	-0.941	-0.928
0.40	-0.401	-0.499	-0.601	-0.944	-0.734
0.45	-0.419	-0.498	-0.603	-0.753	-0.531
0.50	-0.421	-0.496	-0.598	-0.701	-0.530
0.55	-0.430	-0.498	-0.593	-0.700	-0.535
0.60	-0.429	-0.499	-0.585	-0.661	-0.528
0.65	-0.402	-0.498	-0.583	-0.568	-0.523
0.70	-0.452	-0.489	-0.578	-0.526	-0.521
0.75	-0.463	-0.506	-0.558	-0.514	-0.519
0.80	-0.477	-0.522	-0.574	-0.496	-0.521
0.85	-0.458	-0.530	-0.471	-0.446	-0.507
0.90	-0.297	-0.379	-0.161	-0.375	-0.488
0.95	-0.116	-0.088	-0.055	-0.283	-0.473
1.00	-0.009	0.008	-0.007	-0.202	-0.391

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR									
10.35	0.247	0.029			-0.308					
11.35	0.270	0.102			-0.322					
12.35	0.294				-0.338		-0.284			
14.35	0.342	0.089			-0.346					
15.35	0.366	0.077			-0.345				-0.291	
16.35	0.390	0.063			-0.354				0.018	
17.35	0.413	0.049			-0.357		-0.325		-0.282	
18.35	0.437	0.035			-0.365				-0.295	
19.35	0.461	0.029			-0.379				-0.306	
20.35	0.485	0.030			-0.382				-0.312	
22.35	0.533	0.020			-0.386				-0.327	
23.35	0.556	0.024			-0.395				-0.324	
24.35	0.580	0.028			-0.397		-0.369		-0.338	
25.35	0.604	0.029			-0.396				-0.344	
26.35	0.628	0.044			-0.400				-0.350	
27.35	0.652	0.065			-0.396					
30.35	0.723								-0.355	
31.35	0.747		0.136	-0.436		-0.428	-0.390	-0.401	-0.362	
32.35	0.771		0.148	-0.430		-0.428		-0.402	-0.361	
33.35	0.795		0.152	-0.453		-0.438		-0.405	-0.365	
34.35	0.818		0.144	-0.447		-0.433		-0.418	-0.379	
35.35	0.842		0.144	-0.433		-0.416		-0.412	-0.380	
36.35	0.866		0.136	-0.295		-0.410		-0.399	-0.385	
37.35	0.890		0.119	-0.243		-0.341		-0.354	-0.354	
38.35	0.914		0.099	-0.202		-0.263		-0.288	-0.270	
39.35	0.938		0.084	-0.124	-0.073	-0.191		-0.235	-0.162	
40.35	0.961		0.063	-0.066	-0.044	-0.143		-0.166	-0.231	
41.35	0.985		0.028	-0.012	-0.033	-0.069		-0.123	-0.128	
42.35	1.009		0.038	0.006	-0.031	-0.039		-0.077	-0.175	
44.85	1.069		0.040	0.025	0.008	0.004		-0.036	-0.086	
45.85	1.092		0.042	0.035	0.032					
46.85	1.116		0.045	0.036	0.022					

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR								
16.75	0.399			-0.344	-0.318	-0.335			
13.75	0.328		-0.249	-0.304	-0.343	-0.354			
10.75	0.256		-0.282	-0.338	-0.362	-0.385	-0.128	-0.175	
7.75	0.185	-0.284	-0.325	-0.369	-0.390	-0.394	-0.139		-0.050
6.75	0.161				-0.401	-0.399	-0.128	-0.077	-0.036
5.75	0.137	-0.315	-0.348	-0.394	-0.412	-0.413	-0.106	-0.069	-0.028
4.75	0.113				-0.419	-0.402	-0.084	-0.044	-0.030
4.25	0.101	-0.338	-0.357	-0.397					
3.75	0.089				-0.428	-0.410	-0.069	-0.039	0.004
2.75	0.06				-0.434	-0.388	-0.064	-0.022	-0.014
1.75	0.042				-0.442	-0.337	-0.053	-0.027	0.005
0.75	0.018				-0.436	-0.295	-0.033	-0.031	0.008
-0.25	-0.006						-0.012	0.006	0.025
-1.25	-0.030				0.136	0.136	0.028	0.038	0.040
-2.25	-0.054				0.126	0.133	0.048	0.063	0.046
-2.75	-0.066		0.049	0.028					
-3.25	-0.077				0.102	0.121	0.057	0.057	0.049
-4.25	-0.101	0.091	0.047	0.032	0.099	0.119	0.071	0.059	0.037
-5.25	-0.125				0.101	0.101	0.055	0.066	0.052
-6.25	-0.149	0.091	0.064	0.033	0.100	0.093	0.053		-0.043
-9.25	-0.220		0.070	0.047	0.078	0.071	0.062		
-12.25	-0.292		0.077	0.048	0.061	0.065			
-15.25	-0.363			0.053	0.065	0.066			

RUN.SEQ
188.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.921	2.993	6.81	1467	848	546.8	503.3	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.494	0.074	0.567	-0.0636	-0.0096	-0.0732	0.2420	37.88	38.04	37.90	42.66	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.387	0.100	0.487	-0.0804	-0.0193	-0.0997	45.79	44.19	45.46	0.698	0.088
0.296	0.469	0.095	0.564	-0.0672	-0.0259	-0.0931	39.34	52.28	41.52	0.688	-0.049
0.500	0.553	0.080	0.634	-0.0705	-0.0275	-0.0979	37.74	59.19	40.46	0.633	-0.152
0.697	0.659	0.054	0.713	-0.1026	-0.0186	-0.1211	40.58	59.27	42.00	0.561	-0.194
0.894	0.596	-0.040	0.556	-0.1207	0.0013	-0.1194	45.26	28.33	46.49	0.319	-0.123

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.555	0.555	0.555	0.555	0.555
0.01			0.354		
0.02			0.289		
0.03			0.247		
0.04			0.216		
0.05	0.244	0.216	0.187	0.140	0.018
0.10	0.167	0.130	0.094	0.056	-0.050
0.15	0.125	0.089	0.041	0.005	-0.087
0.20	0.097	0.063	0.024	0.001	-0.108
0.30	0.057	0.012	-0.010	-0.028	-0.168
0.40	0.030	-0.009	-0.035	-0.045	-0.184
0.50	-0.011	-0.032	-0.035	-0.066	-0.145
0.55					
0.60	0.018	0.040	0.038	0.020	-0.043
0.65			0.106		
0.70	0.120	0.157	0.159	0.135	0.072
0.75	0.162	0.186	0.197	0.170	0.108
0.80	0.180	0.200	0.205	0.188	0.122
0.85	0.175	0.199	0.200	0.181	0.118
0.90	0.139	0.163	0.173	0.151	0.081
0.95	0.073	0.123	0.122	0.072	-0.009
1.00	0.028	0.069	0.043	-0.123	-0.334

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.547	0.471	0.410	0.376	0.389
0.003			0.068		
0.006			-0.122	-0.130	
0.01	-0.108	-0.269	-0.339	-0.341	-0.247
0.02	-0.400	-0.591	-0.633	-0.597	-0.443
0.03	-0.515	-0.739	-0.778	-0.723	-0.511
0.04	-0.510	-0.823	-0.831	-0.747	-0.506
0.05	-0.535	-0.872	-0.902	-0.887	-0.716
0.06	-0.506	-0.868	-0.929	-0.883	-0.713
0.08	-0.484	-0.863	-0.932	-0.863	-0.674
0.10	-0.465	-0.877	-0.958	-0.888	-0.710
0.125	-0.442	-0.715	-0.957	-0.919	-0.725
0.15	-0.428	-0.668	-0.970	-0.946	-0.748
0.175	-0.427	-0.649	-0.969	-0.947	-0.768
0.20	-0.433	-0.615	-0.935	-0.952	-0.791
0.225	-0.424	-0.576	-0.841	-0.957	-0.824
0.25	-0.409	-0.556	-0.706	-0.954	-0.840
0.30	-0.423	-0.527	-0.634	-0.971	-0.906
0.35	-0.411	-0.504	-0.597	-0.974	-0.944
0.40	-0.420	-0.502	-0.601	-0.901	-0.550
0.45	-0.433	-0.497	-0.606	-0.731	-0.525
0.50	-0.432	-0.499	-0.594	-0.716	-0.524
0.55	-0.437	-0.498	-0.587	-0.710	-0.522
0.60	-0.434	-0.496	-0.579	-0.605	-0.520
0.65	-0.409	-0.481	-0.573	-0.536	-0.519
0.70	-0.467	-0.482	-0.564	-0.505	-0.518
0.75	-0.475	-0.504	-0.555	-0.468	-0.509
0.80	-0.467	-0.470	-0.413	-0.409	-0.504
0.85	-0.371	-0.217	-0.149	-0.341	-0.494
0.90	-0.202	-0.079	-0.044	-0.292	-0.464
0.95	-0.057	0.014	0.012	-0.214	-0.415
1.00	0.028	0.069	0.043	-0.123	-0.334

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.036				-0.319				
11.35	0.270	0.100				-0.333				
12.35	0.294					-0.345		-0.291		
14.35	0.342	0.078				-0.349				
15.35	0.366	0.068				-0.346				-0.303
16.35	0.390	0.064				-0.372				-0.020
17.35	0.413	0.053				-0.375		-0.333		-0.292
18.35	0.437	0.036				-0.381				-0.308
19.35	0.461	0.030				-0.380				-0.309
20.35	0.485	0.035				-0.387				-0.318
22.35	0.533	0.027				-0.393				-0.340
23.35	0.556	0.032				-0.398				-0.328
24.35	0.580	0.048				-0.398		-0.372		-0.342
25.35	0.604	0.041				-0.404				-0.353
26.35	0.628	0.049				-0.392				-0.358
27.35	0.652	0.069				-0.388				
30.35	0.723									-0.360
31.35	0.747		0.149		-0.443		-0.428	-0.401	-0.403	-0.366
32.35	0.771		0.143		-0.434		-0.436		-0.406	-0.358
33.35	0.795		0.174		-0.346		-0.438		-0.408	-0.356
34.35	0.818		0.161		-0.336		-0.403		-0.324	-0.368
35.35	0.842		0.143		-0.308		-0.355		-0.364	-0.330
36.35	0.866		0.136		-0.289		-0.288		-0.357	-0.292
37.35	0.890		0.119		-0.189		-0.166		-0.234	-0.272
38.35	0.914		0.106		-0.114		-0.143		-0.165	-0.252
39.35	0.938		0.096	-0.075	-0.071		-0.096		-0.148	-0.171
40.35	0.961		0.071	-0.029	-0.048		-0.056		-0.092	-0.103
41.35	0.985		0.035	0.018	-0.027		-0.028		-0.082	-0.111
42.35	1.009		0.059	0.053	0.009		0.005		-0.050	-0.145
44.85	1.069		0.051	0.045	0.027		0.023		-0.004	-0.045
45.85	1.092		0.047	0.032	0.037					
46.85	1.116		0.058	0.032	0.039					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.346	-0.324	-0.311			
13.75	0.328		-0.254	-0.311	-0.347	-0.337			
10.75	0.256		-0.292	-0.342	-0.366	-0.292	-0.111	-0.145	
7.75	0.185	-0.291	-0.333	-0.372	-0.401	-0.349	-0.070		0.002
6.75	0.161				-0.403	-0.357	-0.082	-0.050	-0.004
5.75	0.137	-0.324	-0.361	-0.396	-0.422	-0.246	-0.064	-0.027	0.026
4.75	0.113				-0.428	-0.269	-0.036	-0.012	-0.008
4.25	0.101	-0.345	-0.375	-0.398					
3.75	0.089				-0.428	-0.288	-0.028	0.005	0.023
2.75	0.066				-0.423	-0.297	-0.022	0.004	0.022
1.75	0.042				-0.441	-0.302	-0.008	0.013	0.033
0.75	0.018				-0.443	-0.289	-0.027	0.009	0.027
-0.25	-0.006						0.018	0.053	0.045
-1.25	-0.030					0.149	0.136	0.035	0.059	0.051
-2.25	-0.054					0.131	0.132	0.067	0.074	0.050
-2.75	-0.066		0.053	0.048						
-3.25	-0.077					0.122	0.116	0.082	0.064	0.044
-4.25	-0.101	0.095	0.047	0.046		0.113	0.121	0.087	0.066	0.055
-5.25	-0.125					0.105	0.100	0.084	0.066	0.061
-6.25	-0.149	0.101	0.055	0.058		0.109	0.099	0.087		-0.044
-9.25	-0.220		0.071	0.055		0.084	0.090	0.087		
-12.25	-0.292		0.075	0.067		0.070	0.078			
-15.25	-0.363			0.062		0.069	0.074			

RUN.SEQ
189.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.902	2.995	6.81	1479	873	546.8	496.9	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.472	0.085	0.557	-0.0475	-0.0120	-0.0595	0.2380	35.05	39.17	35.67	42.73	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.366	0.113	0.479	-0.0640	-0.0236	-0.0877	42.51	45.93	43.32	0.686	0.108
0.296	0.451	0.104	0.555	-0.0535	-0.0282	-0.0818	36.88	52.08	39.73	0.677	-0.033
0.500	0.528	0.094	0.622	-0.0491	-0.0303	-0.0795	34.31	57.40	37.78	0.621	-0.132
0.697	0.627	0.057	0.684	-0.0757	-0.0208	-0.0966	37.09	61.30	39.12	0.538	-0.174
0.894	0.594	-0.018	0.576	-0.1074	-0.0037	-0.1111	43.07	4.618	44.29	0.331	-0.123

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.553	0.553	0.553	0.553	0.553
0.01			0.377		
0.02			0.306		
0.03			0.258		
0.04			0.225		
0.05	0.251	0.218	0.197	0.152	0.049
0.10	0.178	0.139	0.106	0.068	-0.020
0.15	0.137	0.102	0.059	0.017	-0.054
0.20	0.104	0.073	0.039	-0.011	-0.082
0.30	0.065	0.024	-0.002	-0.042	-0.150
0.40	0.036	0.001	-0.014	-0.056	-0.148
0.50	0.002	-0.018	-0.015	-0.051	-0.125
0.55					
0.60	0.038	0.047	0.063	0.029	-0.031
0.65			0.114		
0.70	0.136	0.165	0.155	0.135	0.083
0.75	0.181	0.200	0.205	0.176	0.113
0.80	0.192	0.212	0.212	0.181	0.138
0.85	0.191	0.211	0.209	0.172	0.132
0.90	0.152	0.173	0.183	0.144	0.093
0.95	0.099	0.124	0.132	0.095	0.017
1.00	0.044	0.069	0.069	-0.010	-0.230

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.527	0.460	0.390	0.353	0.365
0.003			0.036		
0.006			-0.158	-0.174	
0.01	-0.128	-0.305	-0.380	-0.389	-0.300
0.02	-0.417	-0.634	-0.681	-0.648	-0.501
0.03	-0.538	-0.765	-0.834	-0.779	-0.579
0.04	-0.526	-0.871	-0.886	-0.805	-0.551
0.05	-0.540	-0.917	-0.954	-0.949	-0.757
0.06	-0.525	-0.908	-0.983	-0.939	-0.764
0.08	-0.497	-0.898	-0.981	-0.921	-0.739
0.10	-0.475	-0.886	-1.008	-0.940	-0.763
0.125	-0.463	-0.731	-1.008	-0.969	-0.780
0.15	-0.448	-0.692	-1.017	-0.997	-0.799
0.175	-0.444	-0.662	-1.013	-0.994	-0.817
0.20	-0.443	-0.617	-0.967	-0.997	-0.843
0.225	-0.436	-0.581	-0.806	-1.003	-0.871
0.25	-0.421	-0.556	-0.704	-1.000	-0.885
0.30	-0.428	-0.524	-0.643	-1.009	-0.953
0.35	-0.421	-0.506	-0.607	-1.000	-0.917
0.40	-0.431	-0.503	-0.615	-0.706	-0.548
0.45	-0.432	-0.499	-0.610	-0.723	-0.538
0.50	-0.429	-0.498	-0.602	-0.721	-0.535
0.55	-0.436	-0.495	-0.594	-0.696	-0.537
0.60	-0.444	-0.486	-0.587	-0.594	-0.534
0.65	-0.422	-0.483	-0.578	-0.512	-0.520
0.70	-0.462	-0.485	-0.535	-0.412	-0.516
0.75	-0.425	-0.457	-0.252	-0.344	-0.501
0.80	-0.299	-0.188	-0.124	-0.285	-0.456
0.85	-0.179	-0.089	-0.069	-0.224	-0.409
0.90	-0.112	-0.092	-0.021	-0.152	-0.377
0.95	-0.019	0.023	0.033	-0.079	-0.313
1.00	0.044	0.069	0.069	-0.010	-0.230

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.033				-0.333				
11.35	0.270	0.094				-0.349				
12.35	0.294					-0.361		-0.295		
14.35	0.342	0.079				-0.365				
15.35	0.366	0.068				-0.362				-0.306
16.35	0.390	0.061				-0.372				-0.052
17.35	0.413	0.053				-0.369		-0.342		-0.298
18.35	0.437	0.033				-0.376				-0.307
19.35	0.461	0.035				-0.387				-0.311
20.35	0.485	0.029				-0.394				-0.314
22.35	0.533	0.031				-0.399				-0.332
23.35	0.556	0.032				-0.411				-0.331
24.35	0.580	0.052				-0.406		-0.374		-0.343
25.35	0.604	0.058				-0.411				-0.354
26.35	0.628	0.074				-0.393				-0.356
27.35	0.652	0.069				-0.390				
30.35	0.723									-0.369
31.35	0.747		0.142		-0.371		-0.410	-0.389	-0.366	-0.311
32.35	0.771		0.157		-0.361		-0.341		-0.331	-0.302
33.35	0.795		0.164		-0.336		-0.320		-0.335	-0.312
34.35	0.818		0.158		-0.281		-0.296		-0.306	-0.261
35.35	0.842		0.145		-0.235		-0.212		-0.198	-0.194
36.35	0.866		0.143		-0.176		-0.150		-0.159	-0.197
37.35	0.890		0.120		-0.126		-0.128		-0.150	-0.153
38.35	0.914		0.104		-0.086		-0.115		-0.116	-0.109
39.35	0.938		0.101	-0.031	-0.032		-0.068		-0.099	-0.085
40.35	0.961		0.078	-0.000	-0.000		-0.042		-0.061	-0.094
41.35	0.985		0.056	0.035	0.015		-0.024		-0.035	-0.082
42.35	1.009		0.068	0.034	0.031		0.003		-0.019	-0.118
44.85	1.069		0.052	0.051	0.037		0.028		-0.010	-0.035
45.85	1.092		0.051	0.034	0.040					
46.85	1.116		0.045	0.046	0.047					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.349	-0.278	-0.169			
13.75	0.328		-0.265	-0.304	-0.305	-0.180			
10.75	0.256		-0.298	-0.343	-0.311	-0.197	-0.082	-0.118	
7.75	0.185	-0.295	-0.342	-0.374	-0.389	-0.171	-0.057		-0.004
6.75	0.161				-0.366	-0.159	-0.035	-0.019	-0.010
5.75	0.137	-0.329	-0.368	-0.399	-0.392	-0.143	-0.023	-0.028	-0.004
4.75	0.113				-0.364	-0.145	-0.021	-0.017	-0.001
4.25	0.101	-0.361	-0.369	-0.406					
3.75	0.089				-0.410	-0.150	-0.024	0.003	0.028
2.75	0.066				-0.361	-0.151	0.005	0.002	0.025
1.75	0.042				-0.383	-0.169	0.005	0.011	0.048
0.75	0.018				-0.371	-0.176	0.015	0.031	0.037
-0.25	-0.006						0.035	0.034	0.051
-1.25	-0.030					0.142	0.143	0.056	0.068	0.052
-2.25	-0.054					0.131	0.135	0.082	0.096	0.061
-2.75	-0.066			0.053	0.052					
-3.25	-0.077					0.126	0.129	0.069	0.082	0.065
-4.25	-0.101	0.087	0.064	0.043	0.112	0.121	0.084	0.078	0.069
-5.25	-0.125				0.105	0.100	0.086	0.081	0.072
-6.25	-0.149	0.100	0.065	0.064	0.107	0.092	0.081		-0.038
-9.25	-0.220		0.075	0.074	0.083	0.100	0.082		
-12.25	-0.292		0.065	0.083	0.078	0.090			
-15.25	-0.363			0.077	0.073	0.081			

RUN.SEQ
190.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.883	2.993	6.81	1492	899	547.1	490.1	5.00	17

WING COEFFICIENTS

	CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
	0.461	0.086	0.547	-0.0366	-0.0129	-0.0495	0.2358	32.94	40.05	34.05	43.10	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.350	0.110	0.461	-0.0540	-0.0221	-0.0760	40.41	45.02	41.51	0.660	0.121
0.296	0.449	0.093	0.542	-0.0457	-0.0248	-0.0705	35.17	51.64	38.00	0.662	-0.018
0.500	0.527	0.088	0.615	-0.0411	-0.0284	-0.0695	32.80	57.22	36.30	0.615	-0.122
0.697	0.597	0.082	0.679	-0.0506	-0.0275	-0.0781	33.48	58.69	36.51	0.534	-0.162
0.894	0.581	0.001	0.583	-0.0843	-0.0099	-0.0942	39.51	751.4	41.17	0.334	-0.118

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.551	0.551	0.551	0.551	0.551
0.01			0.385		
0.02			0.313		
0.03			0.261		
0.04			0.225		
0.05	0.255	0.207	0.195	0.170	0.071
0.10	0.178	0.122	0.098	0.086	-0.002
0.15	0.135	0.083	0.056	0.040	-0.040
0.20	0.105	0.057	0.033	0.016	-0.066
0.30	0.068	0.017	-0.004	-0.011	-0.133
0.40	0.036	0.001	-0.025	-0.026	-0.129
0.50	0.002	-0.018	-0.022	-0.028	-0.102
0.55					
0.60	0.029	0.038	0.065	0.048	-0.020
0.65			0.122		
0.70	0.123	0.157	0.164	0.153	0.095
0.75	0.172	0.191	0.191	0.187	0.126
0.80	0.184	0.186	0.201	0.200	0.156
0.85	0.179	0.188	0.200	0.203	0.151
0.90	0.146	0.153	0.167	0.173	0.123
0.95	0.106	0.105	0.121	0.126	0.040
1.00	0.036	0.056	0.060	0.049	-0.122

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.496	0.441	0.362	0.321	0.337
0.003			-0.006		
0.006			-0.209	-0.227	
0.01	-0.168	-0.352	-0.439	-0.453	-0.368
0.02	-0.454	-0.691	-0.748	-0.717	-0.574
0.03	-0.561	-0.870	-0.903	-0.850	-0.657
0.04	-0.553	-0.929	-0.957	-0.878	-0.640
0.05	-0.549	-0.977	-1.032	-1.023	-0.819
0.06	-0.531	-0.963	-1.051	-1.012	-0.824
0.08	-0.512	-0.953	-1.048	-0.998	-0.804
0.10	-0.490	-0.892	-1.071	-1.010	-0.823
0.125	-0.466	-0.745	-1.070	-1.033	-0.842
0.15	-0.458	-0.727	-1.075	-1.063	-0.852
0.175	-0.454	-0.689	-1.063	-1.054	-0.877
0.20	-0.457	-0.645	-0.971	-1.058	-0.901
0.225	-0.446	-0.610	-0.779	-1.066	-0.927
0.25	-0.429	-0.583	-0.707	-1.058	-0.939
0.30	-0.431	-0.546	-0.658	-1.054	-1.006
0.35	-0.420	-0.519	-0.628	-0.849	-0.900
0.40	-0.429	-0.511	-0.634	-0.740	-0.565
0.45	-0.435	-0.501	-0.626	-0.746	-0.562
0.50	-0.432	-0.502	-0.613	-0.745	-0.564
0.55	-0.428	-0.495	-0.604	-0.723	-0.564
0.60	-0.420	-0.491	-0.583	-0.546	-0.548
0.65	-0.398	-0.473	-0.466	-0.326	-0.510
0.70	-0.372	-0.440	-0.291	-0.259	-0.462
0.75	-0.301	-0.262	-0.200	-0.226	-0.408
0.80	-0.238	-0.160	-0.133	-0.184	-0.343
0.85	-0.167	-0.108	-0.089	-0.126	-0.284
0.90	-0.080	-0.050	-0.038	-0.070	-0.231
0.95	-0.006	0.006	0.018	-0.005	-0.170
1.00	0.036	0.056	0.060	0.049	-0.122

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.027			-0.359				
11.35	0.270	0.101			-0.370				
12.35	0.294				-0.379		-0.309		
14.35	0.342	0.083			-0.386				
15.35	0.366	0.069			-0.383				-0.312
16.35	0.390	0.061			-0.385				-0.084
17.35	0.413	0.062			-0.388		-0.341		-0.318
18.35	0.437	0.043			-0.400				-0.316
19.35	0.461	0.043			-0.400				-0.304
20.35	0.485	0.033			-0.407				-0.308
22.35	0.533	0.040			-0.404				-0.330
23.35	0.556	0.058			-0.404				-0.325
24.35	0.580	0.049			-0.413		-0.379		-0.341
25.35	0.604	0.048			-0.423				-0.347
26.35	0.628	0.062			-0.394				-0.354
27.35	0.652	0.070			-0.382				
30.35	0.723								-0.316
31.35	0.747		0.146	-0.357		-0.287	-0.261	-0.277	-0.304
32.35	0.771		0.155	-0.352		-0.312		-0.269	-0.236
33.35	0.795		0.152	-0.274		-0.251		-0.204	-0.209
34.35	0.818		0.144	-0.202		-0.212		-0.220	-0.195
35.35	0.842		0.143	-0.167		-0.176		-0.175	-0.182
36.35	0.866		0.136	-0.142		-0.139		-0.159	-0.142
37.35	0.890		0.125	-0.107		-0.094		-0.115	-0.117
38.35	0.914		0.111	-0.086		-0.080		-0.094	-0.104
39.35	0.938		0.109	-0.020	-0.034	-0.050		-0.080	-0.085
40.35	0.961		0.093	0.001	-0.007	-0.027		-0.058	-0.075
41.35	0.985		0.055	0.036	0.011	-0.018		-0.035	-0.068
42.35	1.009		0.052	0.038	0.022	-0.002		-0.028	-0.125
44.85	1.069		0.040	0.038	0.025	0.019		-0.009	-0.037
45.85	1.092		0.059	0.044	0.031				
46.85	1.116		0.055	0.035	0.021				

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR							
16.75	0.399			-0.341	-0.240	-0.160		
13.75	0.328		-0.263	-0.312	-0.291	-0.166		
10.75	0.256		-0.318	-0.341	-0.304	-0.142	-0.068	-0.125
7.75	0.185	-0.309	-0.341	-0.379	-0.261	-0.160	-0.043	-0.007
6.75	0.161				-0.277	-0.159	-0.035	-0.028 -0.009
5.75	0.137	-0.342	-0.372	-0.400	-0.267	-0.121	-0.012	0.006 0.013
4.75	0.113				-0.264	-0.132	-0.020	0.002 0.011
4.25	0.101	-0.379	-0.388	-0.413				
3.75	0.089				-0.287	-0.139	-0.018	-0.002 0.019
2.75	0.066				-0.294	0.119	0.003	0.012 0.034
1.75	0.042				-0.345	-0.107	0.021	0.023 0.040
0.75	0.018				-0.357	-0.142	0.011	0.022 0.025
-0.25	-0.006						0.036	0.038 0.038
-1.25	-0.030					0.146	0.136	0.055 0.052 0.040
-2.25	-0.054					0.130	0.147	0.087 0.073 0.052
-2.75	-0.066			0.062	0.049			
-3.25	-0.077					0.133	0.137	0.093 0.070 0.060
-4.25	-0.101	0.100	0.050	0.056	0.130	0.119	0.089	0.067 0.068
-5.25	-0.125					0.113	0.102	0.087 0.070 0.065
-6.25	-0.149	0.101	0.062	0.070	0.104	0.094	0.084	-0.044
-9.25	-0.220		0.078	0.072	0.096	0.087	0.091	
-12.25	-0.292		0.084	0.071	0.087	0.074		
-15.25	-0.363			0.078	0.076	0.077		

RUN.SEQ
191.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.861	2.993	6.81	1507	929	547.1	482.2	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.454	0.090	0.544	-0.0270	-0.0146	-0.0416	0.2330	30.95	41.20	32.64	42.81	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.366	0.108	0.474	-0.0537	-0.0228	-0.0765	39.67	46.19	41.15	0.679	0.128
0.296	0.424	0.108	0.532	-0.0335	-0.0280	-0.0615	32.90	50.89	36.55	0.650	-0.007
0.500	0.509	0.089	0.598	-0.0307	-0.0286	-0.0594	31.04	57.10	34.92	0.598	-0.110
0.697	0.584	0.085	0.668	-0.0367	-0.0288	-0.0656	31.29	59.08	34.81	0.526	-0.152
0.894	0.580	0.011	0.591	-0.0612	-0.0142	-0.0754	35.57	151.3	37.76	0.339	-0.114

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.548	0.548	0.548	0.548	0.548
0.01			0.386		
0.02			0.314		
0.03			0.266		
0.04			0.231		
0.05	0.238	0.231	0.199	0.182	0.079
0.10	0.162	0.143	0.101	0.089	0.000
0.15	0.119	0.103	0.057	0.034	-0.039
0.20	0.095	0.077	0.036	0.016	-0.065
0.30	0.060	0.034	-0.004	-0.016	-0.124
0.40	0.033	0.013	-0.018	-0.024	-0.120
0.50	0.012	-0.003	-0.024	-0.025	-0.098
0.55					
0.60	0.042	0.051	0.060	0.050	-0.012
0.65			0.110		
0.70	0.130	0.161	0.150	0.156	0.106
0.75	0.165	0.192	0.198	0.187	0.138
0.80	0.175	0.206	0.207	0.207	0.158
0.85	0.176	0.205	0.205	0.211	0.164
0.90	0.145	0.171	0.171	0.184	0.134
0.95	0.109	0.121	0.121	0.138	0.071
1.00	0.024	0.074	0.076	0.050	-0.043

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.483	0.419	0.335	0.289	0.304
0.003			-0.047		
0.006			-0.255	-0.278	
0.01	-0.202	-0.403	-0.495	-0.512	-0.432
0.02	-0.498	-0.753	-0.811	-0.781	-0.645
0.03	-0.608	-0.930	-0.967	-0.917	-0.733
0.04	-0.596	-0.986	-1.021	-0.944	-0.713
0.05	-0.592	-1.036	-1.099	-1.085	-0.865
0.06	-0.575	-1.018	-1.117	-1.080	-0.882
0.08	-0.549	-1.001	-1.106	-1.062	-0.867
0.10	-0.527	-0.847	-1.129	-1.076	-0.884
0.125	-0.500	-0.766	-1.127	-1.094	-0.902
0.15	-0.491	-0.748	-1.120	-1.124	-0.911
0.175	-0.489	-0.687	-1.094	-1.114	-0.927
0.20	-0.486	-0.636	-0.856	-1.116	-0.951
0.225	-0.474	-0.604	-0.740	-1.119	-0.976
0.25	-0.458	-0.572	-0.686	-1.108	-0.984
0.30	-0.464	-0.543	-0.650	-1.086	-1.049
0.35	-0.453	-0.511	-0.624	-0.770	-1.118
0.40	-0.457	-0.506	-0.624	-0.742	-0.788
0.45	-0.462	-0.489	-0.612	-0.754	-0.643
0.50	-0.456	-0.470	-0.595	-0.735	-0.611
0.55	-0.449	-0.436	-0.578	-0.623	-0.556
0.60	-0.414	-0.385	-0.459	-0.378	-0.480
0.65	-0.373	-0.349	-0.300	-0.253	-0.403
0.70	-0.322	-0.280	-0.234	-0.210	-0.315
0.75	-0.297	-0.224	-0.190	-0.187	-0.264
0.80	-0.242	-0.157	-0.145	-0.147	-0.234
0.85	-0.172	-0.109	-0.087	-0.096	-0.170
0.90	-0.091	-0.053	-0.031	-0.042	-0.117
0.95	-0.021	0.023	0.038	0.016	-0.072
1.00	0.024	0.074	0.076	0.050	-0.043

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR								
10.35	0.247	0.021			-0.377				
11.35	0.270	0.092			-0.380				
12.35	0.294				-0.387		-0.341		
14.35	0.342	0.080			-0.390				
15.35	0.366	0.070			-0.388				-0.344
16.35	0.390	0.059			-0.397				-0.119
17.35	0.413	0.063			-0.407		-0.366		-0.328
18.35	0.437	0.054			-0.399				-0.341
19.35	0.461	0.046			-0.407				-0.334
20.35	0.485	0.047			-0.407				-0.341
22.35	0.533	0.044			-0.390				-0.344
23.35	0.556	0.045			-0.422				-0.336
24.35	0.580	0.048			-0.396		-0.367		-0.338
25.35	0.604	0.053			-0.365				-0.330
26.35	0.628	0.063			-0.349				-0.321
27.35	0.652	0.086			-0.349				
30.35	0.723								-0.242
31.35	0.747		0.153	-0.291	-0.284	-0.252	-0.280	-0.238	
32.35	0.771		0.163	-0.255	-0.241		-0.231	-0.217	
33.35	0.795		0.167	-0.216	-0.207		-0.223	-0.198	
34.35	0.818		0.158	-0.204	-0.167		-0.178	-0.189	
35.35	0.842		0.146	-0.168	-0.177		-0.162	-0.175	
36.35	0.866		0.146	-0.140	-0.160		-0.137	-0.147	
37.35	0.890		0.141	-0.109	-0.131		-0.122	-0.139	
38.35	0.914		0.120	-0.071	-0.096		-0.113	-0.140	
39.35	0.938		0.113	-0.027	-0.035	-0.069	-0.092	-0.109	
40.35	0.961		0.083	-0.004	-0.014	-0.037	-0.073	-0.086	
41.35	0.985		0.063	0.021	0.013	-0.042	-0.060	-0.095	
42.35	1.009		0.082	0.040	0.031	-0.022	-0.019	-0.145	
44.85	1.069		0.066	0.033	0.041	-0.011	-0.029	-0.049	
45.85	1.092		0.068	0.038	0.044				
46.85	1.116		0.060	0.025	0.027				

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR								
16.75	0.399			-0.339	-0.201	-0.168			
13.75	0.328		-0.281	-0.310	-0.216	-0.146			
10.75	0.256		-0.328	-0.338	-0.238	-0.147	-0.095	-0.145	
7.75	0.185	-0.341	-0.366	-0.367	-0.252	-0.137	-0.057		-0.017
6.75	0.161				-0.280	-0.137	-0.060	-0.019	-0.029
5.75	0.137	-0.366	-0.396	-0.389	-0.252	-0.129	-0.042	-0.018	-0.000
4.75	0.113				-0.238	-0.135	-0.017	-0.012	0.017
4.25	0.101	-0.387	-0.407	-0.396					
3.75	0.089				-0.284	-0.160	-0.042	-0.022	-0.011
2.75	0.066				-0.259	-0.137	0.007	0.015	0.031
1.75	0.042				-0.278	-0.117	0.010	0.026	0.044
0.75	0.018				-0.291	-0.140	0.013	0.031	0.041
-0.25	-0.006						0.021	0.040	0.033
-1.25	-0.030					0.153	0.146	0.063	0.082	0.066
-2.25	-0.054					0.143	0.148	0.082	0.107	0.058
-2.75	-0.066			0.063	0.048				
-3.25	-0.077					0.117	0.138	0.077	0.092	0.067
-4.25	-0.101	0.099	0.075	0.056		0.111	0.127	0.076	0.089	0.063
-5.25	-0.125					0.110	0.110	0.089	0.082	0.052
-6.25	-0.149	0.103	0.084	0.071		0.117	0.107	0.091		-0.048
-9.25	-0.220		0.091	0.072		0.083	0.105	0.082		
-12.25	-0.292		0.079	0.073		0.087	0.093			
-15.25	-0.363			0.069		0.072	0.086			

RUN.SEQ
192.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.853	2.996	6.82	1514	941	547.0	479.7	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.449	0.088	0.537	-0.0248	-0.0138	-0.0386	0.2312	30.54	40.63	32.19	43.07	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.351	0.106	0.458	-0.0480	-0.0212	-0.0692	38.66	44.95	40.12	0.656	0.133
0.296	0.434	0.099	0.533	-0.0347	-0.0259	-0.0606	33.01	51.22	36.39	0.650	-0.005
0.500	0.501	0.093	0.595	-0.0283	-0.0285	-0.0568	30.65	55.57	34.56	0.594	-0.107
0.697	0.568	0.081	0.649	-0.0311	-0.0272	-0.0583	30.47	58.60	33.98	0.511	-0.145
0.894	0.590	0.018	0.607	-0.0586	-0.0160	-0.0745	34.94	114.7	37.27	0.349	-0.116

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.547	0.547	0.547	0.547	0.547
0.01			0.399		
0.02			0.324		
0.03			0.276		
0.04			0.246		
0.05	0.242	0.217	0.214	0.179	0.076
0.10	0.162	0.134	0.111	0.090	0.011
0.15	0.118	0.094	0.063	0.034	-0.033
0.20	0.098	0.066	0.041	0.018	-0.062
0.30	0.064	0.018	0.003	-0.014	-0.111
0.40	0.029	0.001	-0.016	-0.031	-0.106
0.50	0.011	-0.010	-0.013	-0.035	-0.091
0.55					
0.60	0.046	0.054	0.059	0.050	-0.001
0.65			0.112		
0.70	0.138	0.152	0.158	0.150	0.110
0.75	0.157	0.186	0.196	0.189	0.139
0.80	0.174	0.193	0.210	0.205	0.159
0.85	0.169	0.189	0.202	0.200	0.156
0.90	0.130	0.159	0.167	0.173	0.141
0.95	0.083	0.118	0.121	0.126	0.094
1.00	0.027	0.050	0.066	0.052	-0.047

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.480	0.402	0.323	0.274	0.291
0.003			-0.064		
0.006			-0.278	-0.303	
0.01	-0.216	-0.423	-0.522	-0.542	-0.459
0.02	-0.514	-0.773	-0.837	-0.812	-0.671
0.03	-0.612	-0.954	-0.995	-0.950	-0.757
0.04	-0.606	-1.013	-1.052	-0.975	-0.738
0.05	-0.599	-1.061	-1.130	-1.119	-0.894
0.06	-0.583	-1.040	-1.147	-1.111	-0.905
0.08	-0.556	-1.026	-1.133	-1.092	-0.892
0.10	-0.534	-0.860	-1.156	-1.101	-0.910
0.125	-0.506	-0.792	-1.149	-1.119	-0.926
0.15	-0.492	-0.764	-1.142	-1.149	-0.931
0.175	-0.488	-0.702	-1.064	-1.136	-0.949
0.20	-0.488	-0.654	-0.804	-1.139	-0.974
0.225	-0.477	-0.620	-0.720	-1.141	-0.998
0.25	-0.459	-0.593	-0.680	-1.125	-1.007
0.30	-0.446	-0.561	-0.649	-0.948	-1.074
0.35	-0.430	-0.531	-0.623	-0.739	-1.146
0.40	-0.442	-0.507	-0.621	-0.738	-0.870
0.45	-0.441	-0.490	-0.580	-0.737	-0.660
0.50	-0.438	-0.471	-0.560	-0.647	-0.632
0.55	-0.406	-0.426	-0.476	-0.487	-0.587
0.60	-0.363	-0.393	-0.388	-0.336	-0.510
0.65	-0.345	-0.320	-0.295	-0.263	-0.395
0.70	-0.313	-0.265	-0.241	-0.222	-0.278
0.75	-0.267	-0.230	-0.194	-0.187	-0.225
0.80	-0.213	-0.182	-0.153	-0.140	-0.176
0.85	-0.151	-0.123	-0.091	-0.088	-0.143
0.90	-0.088	-0.061	-0.042	-0.034	-0.115
0.95	-0.017	0.004	0.019	0.015	-0.079
1.00	0.027	0.050	0.066	0.052	-0.047

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR								
10.35	0.247	0.040			-0.365				
11.35	0.270	0.095			-0.371				
12.35	0.294				-0.374		-0.354		
14.35	0.342	0.093			-0.382				
15.35	0.366	0.074			-0.394				-0.350
16.35	0.390	0.061			-0.399				-0.114
17.35	0.413	0.059			-0.403		-0.372		-0.329
18.35	0.437	0.053			-0.417				-0.339
19.35	0.461	0.060			-0.403				-0.322
20.35	0.485	0.057			-0.401				-0.330
22.35	0.533	0.040			-0.390				-0.347
23.35	0.556	0.044			-0.353				-0.338
24.35	0.580	0.058			-0.330		-0.382		-0.330
25.35	0.604	0.059			-0.335				-0.334
26.35	0.628	0.069			-0.348				-0.363
27.35	0.652	0.090			-0.327				
30.35	0.723								-0.276
31.35	0.747		0.152	-0.266		-0.260	-0.284	-0.240	-0.255
32.35	0.771		0.165	-0.236		-0.251		-0.236	-0.222
33.35	0.795		0.169	-0.187		-0.225		-0.208	-0.228
34.35	0.818		0.169	-0.158		-0.169		-0.181	-0.196
35.35	0.842		0.153	-0.128		-0.152		-0.168	-0.173
36.35	0.866		0.157	-0.108		-0.138		-0.150	-0.159
37.35	0.890		0.142	-0.080		-0.116		-0.135	-0.102
38.35	0.914		0.127	-0.045		-0.090		-0.117	-0.094
39.35	0.938		0.123	-0.013	-0.022	-0.077		-0.090	-0.089
40.35	0.961		0.107	0.011	-0.011	-0.048		-0.063	-0.095
41.35	0.985		0.094	0.038	0.014	-0.023		-0.053	-0.097
42.35	1.009		0.080	0.056	0.037	-0.000		-0.047	-0.145
44.85	1.069		0.069	0.045	0.032	0.013		-0.047	-0.049
45.85	1.092		0.064	0.026	0.036				
46.85	1.116		0.053	0.019	0.041				

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR								
16.75	0.399			-0.360	-0.225	-0.173			
13.75	0.328		-0.289	-0.313	-0.234	-0.170			
10.75	0.256		-0.329	-0.330	-0.255	-0.159	-0.097	-0.145	
7.75	0.185	-0.354	-0.372	-0.382	-0.284	-0.153	-0.066		-0.037
6.75	0.161				-0.240	-0.150	-0.053	-0.047	-0.047
5.75	0.137	-0.372	-0.403	-0.425	-0.301	-0.156	-0.047	-0.023	-0.009
4.75	0.113				-0.278	-0.135	-0.053	-0.027	0.009
4.25	0.101	-0.374	-0.403	-0.330					
3.75	0.089				-0.260	-0.138	-0.023	-0.000	0.013
2.75	0.066				-0.284	-0.137	-0.031	-0.009	0.017
1.75	0.042				-0.274	-0.136	-0.014	0.000	0.018
0.75	0.018				-0.266	-0.108	0.014	0.037	0.032
-0.25	-0.006						0.038	0.056	0.045
-1.25	-0.030				0.152	0.157	0.094	0.080	0.069
-2.25	-0.054				0.133	0.134	0.085	0.105	0.076
-2.75	-0.066			0.059	0.058				
-3.25	-0.077				0.142	0.134	0.088	0.084	0.069
-4.25	-0.101	0.108	0.078	0.066	0.136	0.133	0.105	0.090	0.057
-5.25	-0.125				0.117	0.122	0.087	0.091	0.056
-6.25	-0.149	0.112	0.082	0.073	0.118	0.118	0.087		-0.041
-9.25	-0.220		0.080	0.090	0.108	0.097	0 096		
-12.25	-0.292		0.090	0.097	0.091	0.086			
-15.25	-0.363			0.090	0.074	0.078			

RUN.SEQ
193.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.843	2.994	6.81	1522	955	547.1	475.7	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.448	0.087	0.535	-0.0235	-0.0137	-0.0372	0.2300	30 24	40.69	31.95	42.99	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.347	0.108	0.454	-0.0450	-0.0227	-0.0676	37.96	46.08	39.89	0.651	0.134
0.296	0.436	0.099	0.536	-0.0344	-0.0257	-0.0602	32.89	50.92	36.23	0.653	-0.004
0.500	0.504	0.093	0.597	-0.0285	-0.0287	-0.0572	30.66	55.85	34.58	0.597	-0.108
0.697	0.563	0.076	0.640	-0.0298	-0.0258	-0.0556	30.30	58.72	33.69	0.503	-0.141
0.894	0.584	0.014	0.598	-0.0522	-0.0153	-0.0675	33.94	135.0	36.29	0.343	-0.112

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.546	0.546	0.546	0.546	0.546
0.01			0.384		
0.02			0.311		
0.03			0.263		
0.04			0.226		
0.05	0.240	0.223	0.199	0.174	0.092
0.10	0.163	0.132	0.103	0.086	0.001
0.15	0.122	0.085	0.071	0.030	-0.045
0.20	0.094	0.062	0.050	0.011	-0.067
0.30	0.055	0.027	0.008	-0.021	-0.122
0.40	0.026	0.011	-0.014	-0.036	-0.119
0.50	0.024	-0.005	-0.018	-0.023	-0.090
0.55					
0.60	0.051	0.038	0.054	0.040	-0.015
0.65			0.110		
0.70	0.128	0.144	0.153	0.150	0.106
0.75	0.158	0.184	0.194	0.179	0.139
0.80	0.171	0.202	0.209	0.190	0.157
0.85	0.171	0.194	0.202	0.188	0.156
0.90	0.146	0.155	0.175	0.166	0.136
0.95	0.100	0.118	0.130	0.122	0.096
1.00	0.041	0.056	0.051	0.049	-0.021

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.469	0.399	0.311	0.264	0.275
0.003			-0.084		
0.006			-0.301	-0.328	
0.01	-0.226	-0.445	-0.547	-0.572	-0.489
0.02	-0.525	-0.799	-0.868	-0.841	-0.706
0.03	-0.621	-0.982	-1.029	-0.984	-0.791
0.04	-0.612	-1.042	-1.084	-1.010	-0.774
0.05	-0.607	-1.089	-1.164	-1.161	-0.927
0.06	-0.595	-1.057	-1.179	-1.142	-0.932
0.08	-0.565	-1.049	-1.165	-1.131	-0.921
0.10	-0.540	-0.862	-1.186	-1.132	-0.937
0.125	-0.513	-0.804	-1.178	-1.148	-0.951
0.15	-0.503	-0.777	-1.171	-1.179	-0.956
0.175	-0.498	-0.708	-1.022	-1.167	-0.976
0.20	-0.483	-0.663	-0.802	-1.167	-1.000
0.225	-0.475	-0.629	-0.719	-1.163	-1.019
0.25	-0.456	-0.589	-0.676	-1.129	-1.025
0.30	-0.450	-0.557	-0.652	-0.826	-1.087
0.35	-0.442	-0.522	-0.617	-0.730	-1.167
0.40	-0.436	-0.514	-0.595	-0.748	-0.957
0.45	-0.444	-0.492	-0.585	-0.656	-0.658
0.50	-0.415	-0.456	-0.527	-0.553	-0.578
0.55	-0.375	-0.428	-0.456	-0.436	-0.500
0.60	-0.357	-0.382	-0.375	-0.347	-0.395
0.65	-0.329	-0.326	-0.312	-0.279	-0.317
0.70	-0.297	-0.268	-0.252	-0.230	-0.242
0.75	-0.250	-0.230	-0.208	-0.201	-0.211
0.80	-0.204	-0.186	-0.167	-0.159	-0.186
0.85	-0.148	-0.127	-0.104	-0.102	-0.157
0.90	-0.083	-0.065	-0.043	-0.048	-0.123
0.95	-0.012	0.001	0.016	0.008	-0.077
1.00	0.041	0.056	0.051	0.049	-0.021

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.038				-0.395				
11.35	0.270	0.111				-0.394				
12.35	0.294					-0.402		-0.323		
14.35	0.342	0.082				-0.400				
15.35	0.366	0.067				-0.385				-0.331
16.35	0.390	0.062				-0.407				-0.107
17.35	0.413	0.060				-0.415		-0.348		-0.302
18.35	0.437	0.063				-0.403				-0.301
19.35	0.461	0.066				-0.396				-0.326
20.35	0.485	0.042				-0.380				-0.322
22.35	0.533	0.044				-0.367				-0.311
23.35	0.556	0.047				-0.371				-0.300
24.35	0.580	0.061				-0.370		-0.350		-0.295
25.35	0.604	0.069				-0.369				-0.291
26.35	0.628	0.095				-0.307				-0.291
27.35	0.652	0.079				-0.315				
30.35	0.723									-0.236
31.35	0.747		0.133		-0.248		-0.235	-0.234	-0.233	-0.216
32.35	0.771		0.145		-0.213		-0.237		-0.211	-0.210
33.35	0.795		0.163		-0.191		-0.216		-0.183	-0.171
34.35	0.818		0.161		-0.184		-0.178		-0.181	-0.174
35.35	0.842		0.151		-0.146		-0.145		-0.153	-0.157
36.35	0.866		0.140		-0.114		-0.137		-0.131	-0.130
37.35	0.890		0.124		-0.095		-0.105		-0.106	-0.120
38.35	0.914		0.114		-0.054		-0.096		-0.092	-0.115
39.35	0.938		0.115	-0.037	-0.035		-0.073		-0.079	-0.093
40.35	0.961		0.084	-0.011	-0.005		-0.029		-0.057	-0.065
41.35	0.985		0.065	0.019	0.019		-0.032		-0.038	-0.063
42.35	1.009		0.074	0.050	0.043		-0.001		-0.031	-0.130
44.85	1.069		0.056	0.039	0.039		0.019		-0.019	0.063
45.85	1.092		0.053	0.031	0.038					
46.85	1.116		0.049	0.015	0.009					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.340	-0.174	-0.121			
13.75	0.328		-0.262	-0.288	-0.174	-0.118			
10.75	0.256		-0.302	-0.295	-0.216	-0.130	-0.063	-0.130	
7.75	0.185	-0.323	-0.348	-0.350	-0.234	-0.129	-0.046		-0.028
6.75	0.161				-0.233	-0.131	-0.038	-0.031	-0.019
5.75	0.137	-0.373	-0.366	-0.356	-0.221	-0.123	-0.038	-0.000	0.006
4.75	0.113				-0.229	-0.121	-0.051	-0.007	-0.008
4.25	0.101	-0.402	-0.415	-0.370					
3.75	0.089				-0.235	-0.137	-0.032	-0.001	0.019
2.75	0.066				-0.250	-0.109	-0.018	0.017	0.003
1.75	0.042				-0.243	-0.102	0.013	0.025	0.009
0.75	0.018				-0.248	-0.114	0.019	0.043	0.039
-0.25	-0.006						0.019	0.050	0.039
-1.25	-0.030					0.133	0.140	0.065	0.074	0.056
-2.25	-0.054					0.134	0.142	0.091	0.085	0.036
-2.75	-0.066			0.060	0.061					
-3.25	-0.077					0.122	0.122	0.082	0.083	0.041
-4.25	-0.101	0.092	0.053	0.061	0.110	0.121	0.085	0.093	0.050
-5.25	-0.125				0.117	0.101	0.077	0.067	0.048
-6.25	-0.149	0.091	0.067	0.076	0.114	0.102	0.091		-0.050
-9.25	-0.220		0.081	0.074	0.084	0.092	0.088		
-12.25	-0.292		0.082	0.078	0.082	0.069			
-15.25	-0.363			0.068	0.084	0.059			

RUN.SEQ
194.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.833	2.999	6.82	1532	972	546.6	471.9	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.446	0.085	0.531	-0.0219	-0.0136	-0.0356	0.2279	29.92	40.95	31.70	42.93	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.349	0.100	0.449	-0.0444	-0.0196	-0.0639	37.71	44.56	39.24	0.644	0.139
0.296	0.445	0.094	0.540	-0.0369	-0.0251	-0.0620	33.28	51.57	36.48	0.659	-0.006
0.500	0.495	0.097	0.591	-0.0261	-0.0296	-0.0557	30.27	55.70	34.42	0.591	-0.106
0.697	0.553	0.077	0.630	-0.0291	-0.0250	-0.0541	30.26	57.40	33.58	0.496	-0.139
0.894	0.571	0.019	0.590	-0.0481	-0.0159	-0.0640	33.42	109.5	35.85	0.338	-0.110

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.545	0.545	0.545	0.545	.545
0.01			0.396		
0.02			0.319		
0.03			0.273		
0.04			0.245		
0.05	0.239	0.207	0.213	0.186	0.083
0.10	0.163	0.124	0.110	0.090	0.013
0.15	0.122	0.078	0.066	0.033	-0.033
0.20	0.094	0.048	0.041	0.018	-0.059
0.30	0.052	0.018	0.008	-0.013	-0.105
0.40	0.024	0.009	-0.012	-0.027	-0.106
0.50	0.001	-0.003	-0.005	-0.037	-0.096
0.55					
0.60	0.038	0.050	0.070	0.034	-0.004
0.65			0.124		
0.70	0.128	0.152	0.161	0.138	0.108
0.75	0.151	0.177	0.185	0.177	0.141
0.80	0.159	0.183	0.207	0.195	0.158
0.85	0.155	0.187	0.204	0.191	0.161
0.90	0.136	0.155	0.176	0.167	0.137
0.95	0.083	0.108	0.130	0.123	0.094
1.00	0.037	0.039	0.062	0.046	-0.030

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.455	0.376	0.291	0.244	0.259
0.003			-0.114		
0.006			-0.326	-0.358	
0.01	-0.243	-0.476	-0.578	-0.605	-0.520
0.02	-0.544	-0.833	-0.903	-0.888	-0.741
0.03	-0.640	-0.964	-1.061	-1.022	-0.827
0.04	-0.631	-1.071	-1.122	-1.049	-0.813
0.05	-0.628	-1.118	-1.211	-1.193	-0.956
0.06	-0.599	-1.085	-1.218	-1.181	-0.970
0.08	-0.574	-1.078	-1.199	-1.164	-0.952
0.10	-0.551	-0.871	-1.215	-1.177	-0.970
0.125	-0.527	-0.829	-1.208	-1.177	-0.980
0.15	-0.518	-0.782	-1.197	-1.206	-0.986
0.175	-0.513	-0.717	-0.920	-1.190	-0.995
0.20	-0.498	-0.671	-0.766	-1.189	-1.019
0.225	-0.493	-0.619	-0.710	-1.173	-1.036
0.25	-0.465	-0.594	-0.667	-0.990	-1.043
0.30	-0.455	-0.567	-0.619	-0.714	-1.101
0.35	-0.430	-0.521	-0.581	-0.689	-1.171
0.40	-0.424	-0.507	-0.571	-0.671	-0.769
0.45	-0.429	-0.477	-0.534	-0.622	-0.569
0.50	-0.406	-0.477	-0.479	-0.527	-0.496
0.55	-0.380	-0.444	-0.416	-0.451	-0.379
0.60	-0.369	-0.387	-0.367	-0.340	-0.337
0.65	-0.320	-0.316	-0.310	-0.291	-0.298
0.70	-0.288	-0.272	-0.270	-0.247	-0.261
0.75	-0.260	-0.239	-0.213	-0.205	-0.246
0.80	-0.208	-0.183	-0.165	-0.163	-0.208
0.85	-0.153	-0.122	-0.105	-0.114	-0.170
0.90	-0.079	-0.119	-0.052	-0.057	-0.136
0.95	-0.010	-0.016	0.019	-0.001	-0.087
1.00	0.037	0.039	0.062	0.046	-0.030

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.027				-0.389				
11.35	0.270	0.107				-0.384				
12.35	0.294					-0.405		-0.347		
14.35	0.342	0.071				-0.402				
15.35	0.366	0.058				-0.395				-0.335
16.35	0.390	0.051				-0.388				-0.116
17.35	0.413	0.041				-0.383		-0.353		-0.333
18.35	0.437	0.028				-0.402				-0.325
19.35	0.461	0.032				-0.388				-0.315
20.35	0.485	0.036				-0.393				-0.308
22.35	0.533	0.043				-0.382				-0.320
23.35	0.556	0.039				-0.367				-0.328
24.35	0.580	0.046				-0.364		-0.337		-0.328
25.35	0.604	0.053				-0.360				-0.312
26.35	0.628	0.070				-0.327				-0.285
27.35	0.652	0.072				-0.330				
30.35	0.723									-0.234
31.35	0.747		0.138	-0.255		-0.256	-0.233	-0.232	-0.251	
32.35	0.771		0.149	-0.219		-0.223	-0.234	-0.228		
33.35	0.795		0.155	-0.193		-0.217	-0.215	-0.199		
34.35	0.818		0.145	-0.176		-0.182	-0.167	-0.165		
35.35	0.842		0.148	-0.147		-0.160	-0.159	-0.147		
36.35	0.866		0.129	-0.134		-0.146	-0.142	-0.148		
37.35	0.890		0.116	-0.106		-0.122	-0.113	-0.116		
38.35	0.914		0.103	-0.077		-0.079	-0.110	-0.108		
39.35	0.938		0.096	-0.017	-0.048	-0.070	-0.088	-0.111		
40.35	0.961		0.076	-0.011	-0.013	-0.055	-0.075	-0.100		
41.35	0.985		0.059	0.021	0.017	-0.041	-0.060	-0.078		
42.35	1.009		0.071	0.035	0.032	-0.031	-0.039	-0.140		
44.85	1.069		0.054	0.031	0.042	-0.020	-0.025	-0.071		
45.85	1.092		0.054	0.016	0.038					
46.85	1.116		0.037	0.017	0.017					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.313	-0.193	-0.150			
13.75	0.328		-0.273	-0.276	-0.202	-0.134			
10.75	0.256		-0.333	-0.328	-0.251	-0.148	-0.078	-0.140	
7.75	0.185	-0.347	-0.353	-0.337	-0.233	-0.143	-0.073		-0.030
6.75	0.161				-0.232	-0.142	-0.060	-0.039	-0.025
5.75	0.137	-0.384	-0.390	-0.363	-0.266	-0.143	-0.057	-0.050	-0.009
4.75	0.113				-0.261	-0.148	-0.042	-0.040	-0.012
4.25	0.101	-0.405	-0.383	-0.364					
3.75	0.089				-0.256	-0.146	-0.041	-0.031	-0.020
2.75	0.066				-0.237	-0.153	-0.018	-0.007	-0.001
1.75	0.042				-0.245	-0.130	-0.011	0.013	0.011
0.75	0.018				-0.255	-0.134	0.017	0.032	0.042
-0.25	-0.006						0.021	0.035	0.031
-1.25	-0.030				0.138	0.129	0.059	0.071	0.054
-2.25	-0.054				0.120	0.128	0.082	0.082	0.070
-2.75	-0.066		0.041	0.046					
-3.25	-0.077				0.119	0.115	0.075	0.073	0.053
-4.25	-0.101	0.086	0.065	0.050	0.112	0.117	0.078	0.081	0.047
-5.25	-0.125				0.107	0.110	0.079	0.092	0.054
-6.25	-0.149	0.093	0.071	0.063	0.088	0.096	0.081		-0.044
-9.25	-0.220		0.075	0.054	0.071	0.080	0.092		
-12.25	-0.292		0.069	0.075	0.075	0.084			
-15.25	-0.363			0.087	0.073	0.079			

RUN.SEQ
195.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.822	2.999	6.82	1541	989	546.8	467.9	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.443	0.086	0.527	-0.0207	-0.0138	-0.0345	0.2257	29.70	41.06	31.55	42.79	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.348	0.100	0.448	-0.0432	-0.0202	-0.0634	37.40	45.32	39.16	0.642	0.139
0.296	0.443	0.091	0.535	-0.0372	-0.0233	-0.0605	33.39	50.54	36.32	0.652	-0.005
0.500	0.501	0.092	0.593	-0.0271	-0.0275	-0.0545	30.41	54.77	34.20	0.593	-0.105
0.697	0.540	0.087	0.627	-0.0280	-0.0270	-0.0550	30.18	55.92	33.77	0.494	-0.139
0.894	0.544	0.024	0.568	-0.0450	-0.0161	-0.0610	33.28	50.82	35.75	0.326	-0.105

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.544	0.544	0.544	0.544	0.544
0.01			0.395		
0.02			0.319		
0.03			0.272		
0.04			0.235		
0.05	0.225	0.218	0.205	0.190	0.106
0.10	0.154	0.128	0.107	0.099	0.011
0.15	0.120	0.080	0.075	0.044	-0.030
0.20	0.085	0.044	0.048	0.028	-0.056
0.30	0.054	0.024	0.000	0.005	-0.089
0.40	0.023	0.010	-0.009	-0.005	-0.089
0.50	0.013	-0.011	-0.013	-0.025	-0.081
0.55					
0.60	0.039	0.028	0.063	0.049	-0.011
0.65			0.116		
0.70	0.123	0.139	0.153	0..43	0.104
0.75	0.165	0.169	0.183	0.179	0.142
0.80	0.176	0.191	0.201	0.197	0.154
0.85	0.157	0.187	0.198	0.197	0.164
0.90	0.128	0.152	0.164	0.172	0.140
0.95	0.079	0.100	0.114	0.132	0.085
1.00	0.020	0.047	0.056	0.046	-0.004

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.453	0.370	0.275	0.223	0.243
0.003			-0.135		
0.006			-0.356	-0.386	
0.01	-0.266	-0.504	-0.609	-0.639	-0.550
0.02	-0.562	-0.863	-0.940	-0.923	-0.772
0.03	-0.652	-0.963	-1.098	-1.059	-0.859
0.04	-0.645	-1.101	-1.155	-1.085	-0.838
0.05	-0.645	-1.144	-1.243	-1.232	-0.987
0.06	-0.613	-1.105	-1.250	-1.213	-0.999
0.08	-0.589	-1.086	-1.227	-1.193	-0.966
0.10	-0.558	-0.888	-1.247	-1.205	-0.981
0.125	-0.527	-0.894	-1.237	-1.206	-0.985
0.15	-0.520	-0.780	-1.208	-1.233	-0.981
0.175	-0.513	-0.718	-0.844	-1.212	-0.992
0.20	-0.505	-0.668	-0.754	-1.206	-1.009
0.225	-0.490	-0.607	-0.702	-0.940	-1.013
0.25	-0.461	-0.569	-0.668	-0.755	-1.010
0.30	-0.464	-0.546	-0.635	-0.674	-0.875
0.35	-0.440	-0.504	-0.583	-0.659	-0.812
0.40	-0.439	-0.492	-0.555	-0.641	-0.730
0.45	-0.414	-0.475	-0.530	-0.586	-0.608
0.50	-0.390	-0.423	-0.481	-0.527	-0.484
0.55	-0.378	-0.399	-0.438	-0.464	-0.417
0.60	-0.341	-0.361	-0.366	-0.373	-0.366
0.65	-0.306	-0.318	-0.315	-0.307	-0.313
0.70	-0.283	-0.275	-0.283	-0.258	-0.274
0.75	-0.254	-0.244	-0.218	-0.210	-0.247
0.80	-0.200	-0.192	-0.176	-0.163	-0.206
0.85	-0.149	-0.142	-0.119	-0.114	-0.163
0.90	-0.085	-0.197	-0.053	-0.049	-0.119
0.95	-0.017	-0.004	0.018	0.012	-0.069
1.00	0.020	0.047	0.056	0.046	-0.004

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.016				-0.397				
11.35	0.270	0.088				-0.397				
12.35	0.294					-0.415		-0.355		
14.35	0.342	0.071				-0.412				
15.35	0.366	0.054				-0.403				-0.344
16.35	0.390	0.038				-0.410				-0.105
17.35	0.413	0.057				-0.393		-0.357		-0.297
18.35	0.437	0.049				-0.386				-0.300
19.35	0.461	0.048				-0.371				-0.303
20.35	0.485	0.032				-0.370				-0.313
22.35	0.533	0.029				-0.366				-0.308
23.35	0.556	0.036				-0.357				-0.287
24.35	0.580	0.038				-0.352		-0.297		-0.289
25.35	0.604	0.041				-0.346				-0.277
26.35	0.628	0.063				-0.308				-0.277
27.35	0.652	0.078				-0.314				
30.35	0.723									-0.224
31.35	0.747		0.132		-0.245		-0.246	-0.213	-0.238	-0.199
32.35	0.771		0.128		-0.214		-0.239		-0.208	-0.196
33.35	0.795		0.137		-0.204		-0.218		-0.190	-0.185
34.35	0.818		0.135		-0.175		-0.169		-0.157	-0.172
35.35	0.842		0.136		-0.168		-0.156		-0.129	-0.167
36.35	0.866		0.128		-0.124		-0.134		-0.121	-0.144
37.35	0.890		0.115		-0.100		-0.110		-0.098	-0.129
38.35	0.914		0.095		-0.066		-0.082		-0.087	-0.119
39.35	0.938		0.105	-0.043	-0.027		-0.079		-0.088	-0.096
40.35	0.961		0.073	-0.014	-0.017		-0.056		-0.075	-0.094
41.35	0.985		0.051	0.024	0.002		-0.035		-0.056	-0.084
42.35	1.009		0.047	0.026	0.016		-0.018		-0.034	-0.131
44.85	1.069		0.023	0.012	0.015		-0.001		-0.043	-0.065
45.85	1.092		0.029	0.022	0.014					
46.85	1.116		0.038	0.012	0.021					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.313	-0.171	-0.152			
13.75	0.328		-0.283	-0.261	-0.172	-0.152			
10.75	0.256		-0.297	-0.289	-0.199	-0.144	-0.084	-0.131	
7.75	0.185	-0.355	-0.357	-0.297	-0.213	-0.138	-0.067		-0.052
6.75	0.161				-0.238	-0.121	-0.056	-0.034	-0.043
5.75	0.137	-0.385	-0.375	-0.305	-0.212	-0.140	-0.059	-0.027	-0.019
4.75	0.113				-0.224	-0.136	-0.044	-0.020	-0.026
4.25	0.101	-0.415	-0.393	-0.352					
3.75	0.089				-0.246	-0.134	-0.035	-0.018	-0.001
2.75	0.066				-0.250	-0.126	-0.020	-0.005	-0.016
1.75	0.042				-0.243	-0.130	-0.005	-0.003	0.020
0.75	0.018				-0.245	-0.124	0.002	0.016	0.015
-0.25	-0.006						0.024	0.026	0.012
-1.25	-0.030				0.132	0.128	0.051	0.047	0.023
-2.25	-0.054				0.116	0.120	0.070	0.070	0.048
-2.75	-0.066		0.057	0.038					
-3.25	-0.077				0.118	0.116	0.066	0.074	0.048
-4.25	-0.101	0.083	0.047	0.056	0.105	0.107	0.074	0.073	0.040
-5.25	-0.125				0.098	0.096	0.069	0.074	0.037
-6.25	-0.149	0.083	0.058	0.065	0.093	0.102	0.072		-0.061
9.25	-0.220		0.057	0.053	0.076	0.068	0.089		
-12.25	-0.292		0.077	0.058	0.066	0.068			
-15.25	-0.363			0.061	0.053	0.066			

RUN.SEQ
196.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.810	2.997	6.82	1551	1007	546.5	462.3	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.426	0.092	0.518	-0.0190	-0.0140	-0.0330	0.2220	29.45	40.27	31.37	42.89	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.336	0.107	0.443	-0.0395	-0.0215	-0.0610	36.78	45.06	38.79	0.635	0.141
0.296	0.409	0.103	0.512	-0.0280	-0.0262	-0.0542	31.84	50.45	35.57	0.625	0.001
0.500	0.487	0.095	0.583	-0.0269	-0.0281	-0.0549	30.51	54.42	34.43	0.582	-0.104
0.697	0.536	0.086	0.622	-0.0289	-0.0264	-0.0553	30.40	55.70	33.89	0.489	-0.138
0.894	0.539	0.029	0.568	-0.0463	-0.0162	-0.0625	33.58	80.74	36.00	0.326	-0.106

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.542	0.542	0.542	0.542	0.542
0.01			0.409		
0.02			0.328		
0.03			0.278		
0.04			0.247		
0.05	0.241	0.226	0.219	0.195	0.104
0.10	0.176	0.138	0.120	0.103	0.030
0.15	0.139	0.098	0.064	0.048	-0.012
0.20	0.093	0.072	0.041	0.029	-0.042
0.30	0.063	0.028	0.006	-0.006	-0.086
0.40	0.033	0.010	-0.001	-0.023	-0.085
0.50	0.011	-0.006	-0.005	-0.016	-0.073
0.55					
0.60	0.042	0.055	0.053	0.052	-0.006
0.65			0.112		
0.70	0.124	0.155	0.153	0.154	0.108
0.75	0.165	0.191	0.183	0.182	0.142
0.80	0.170	0.204	0.205	0.194	0.161
0.85	0.163	0.191	0.206	0.194	0.155
0.90	0.149	0.150	0.171	0.162	0.134
0.95	0.097	0.113	0.121	0.121	0.093
1.00	0.024	0.053	0.047	0.045	-0.019

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.442	0.352	0.261	0.207	0.228
0.003			-0.154		
0.006			-0.380	-0.412	
0.01	-0.291	-0.525	-0.640	-0.673	-0.579
0.02	-0.582	-0.891	-0.974	-0.949	-0.804
0.03	-0.669	-1.066	-1.141	-1.093	-0.888
0.04	-0.660	-1.125	-1.193	-1.119	-0.865
0.05	-0.646	-1.161	-1.268	-1.269	-1.013
0.06	-0.620	-1.101	-1.282	-1.250	-1.018
0.08	-0.589	-0.972	-1.256	-1.226	-1.000
0.10	-0.560	-0.856	-1.267	-1.227	-1.009
0.125	-0.532	-0.784	-1.253	-1.235	-1.006
0.15	-0.518	-0.740	-0.949	-1.255	-0.996
0.175	-0.503	-0.676	-0.778	-1.228	-0.993
0.20	-0.490	-0.630	-0.718	-1.032	-0.975
0.225	-0.477	-0.578	-0.670	-0.738	-0.811
0.25	-0.445	-0.547	-0.633	-0.666	-0.801
0.30	-0.445	-0.512	-0.620	-0.663	-0.768
0.35	-0.417	-0.468	-0.570	-0.652	-0.806
0.40	-0.410	-0.446	-0.535	-0.632	-0.824
0.45	-0.373	-0.425	-0.483	-0.597	-0.773
0.50	-0.361	-0.391	-0.452	-0.537	-0.457
0.55	-0.347	-0.359	-0.421	-0.457	-0.594
0.60	-0.335	-0.335	-0.365	-0.394	-0.356
0.65	-0.297	-0.293	-0.311	-0.327	-0.324
0.70	-0.274	-0.251	-0.266	-0.268	-0.267
0.75	-0.242	-0.216	-0.226	-0.211	-0.237
0.80	-0.196	-0.177	-0.184	-0.160	-0.199
0.85	-0.136	-0.125	-0.124	-0.114	-0.160
0.90	-0.077	-0.066	-0.060	-0.054	-0.125
0.95	-0.011	0.004	0.003	0.004	-0.085
1.00	0.024	0.053	0.047	0.045	-0.019

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.011				-0.388				
11.35	0.270	0.082				-0.373				
12.35	0.294					-0.405		-0.347		
14.35	0.342	0.067				-0.389				
15.35	0.366	0.064				-0.384				-0.329
16.35	0.390	0.064				-0.381				-0.120
17.35	0.413	0.044				-0.371		-0.363		-0.301
18.35	0.437	0.039				-0.364				-0.301
19.35	0.461	0.036				-0.385				-0.295
20.35	0.485	0.037				-0.374				-0.293
22.35	0.533	0.030				-0.332				-0.305
23.35	0.556	0.036				-0.334				-0.297
24.35	0.580	0.056				-0.324		-0.297		-0.284
25.35	0.604	0.062				-0.305				-0.270
26.35	0.628	0.064				-0.291				-0.260
27.35	0.652	0.080				-0.286				
30.35	0.723									-0.227
31.35	0.747		0.130		-0.240		-0.227	-0.219	-0.216	-0.205
32.35	0.771		0.142		-0.204		-0.228		-0.207	-0.194
33.35	0.795		0.150		-0.205		-0.226		-0.178	-0.167
34.35	0.818		0.145		-0.178		-0.184		-0.160	-0.182
35.35	0.842		0.142		-0.150		-0.150		-0.148	-0.160
36.35	0.866		0.140		-0.134		-0.135		-0.133	-0.144
37.35	0.890		0.118		-0.110		-0.122		-0.103	-0.108
38.35	0.914		0.105		-0.067		-0.097		-0.095	-0.098
39.35	0.938		0.089	-0.030	-0.065		-0.064		-0.077	-0.090
40.35	0.961		0.062	-0.015	-0.032		-0.033		-0.061	-0.073
41.35	0.985		0.050	0.019	-0.004		-0.039		-0.047	-0.078
42.35	1.009		0.069	0.030	0.026		-0.032		-0.030	-0.154
44.85	1.069		0.052	0.027	0.020		-0.024		-0.032	-0.073
45.85	1.092		0.052	0.033	0.011					
46.85	1.116		0.037	0.009	0.012					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.95
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.290	-0.197	-0.132			
13.75	0.328		-0.291	-0.250	-0.199	-0.143			
10.75	0.256		-0.301	-0.284	-0.205	-0.144	-0.078	-0.154	
7.75	0.185	-0.347	-0.363	-0.297	-0.219	-0.141	-0.078		-0.047
6.75	0.161				-0.216	-0.133	-0.047	-0.030	-0.032
5.75	0.137	-0.379	-0.395	-0.315	-0.230	-0.129	-0.052	-0.031	-0.033
4.75	0.113				-0.236	-0.133	-0.038	-0.027	-0.015
4.25	0.101	-0.405	-0.371	-0.324					
3.75	0.089				-0.227	-0.135	-0.039	-0.032	-0.024
2.75	0.066				-0.231	-0.128	-0.014	-0.006	0.001
1.75	0.042				-0.242	-0.137	-0.013	0.000	0.012
0.75	0.018				-0.240	-0.134	-0.004	0.026	0.020
-0.25	-0.006						0.019	0.030	0.027
-1.25	-0.030				0.130	0.140	0.050	0.069	0.052
-2.25	-0.054				0.146	0.118	0.074	0.079	0.054
-2.75	-0.066		0.044	0.056					
3.25	-0.077				0.108	0.119	0.066	0.070	0.050
-4.25	-0.101	0.078	0.057	0.045	0.091	0.098	0.067	0.069	0.041
-5.25	-0.125				0.089	0.100	0.077	0.077	0.044
-6.25	-0.149	0.082	0.065	0.048	0.086	0.094	0.065		-0.053
-9.25	-0.220		0.071	0.052	0.081	0.095	0 073		
-12.25	-0.292		0.065	0.058	0.069	0.066			
-15.25	-0.363			0.060	0.061	0.053			

RUN.SEQ
197.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.801	2.996	6.82	1558	1022	546.4	458.5	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.417	0.092	0.509	-0.0173	-0.0145	-0.0317	0.2173	29.15	40.75	31.24	42.72	0.009	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.330	0.107	0.437	-0.0377	-0.0231	-0.0608	36.40	46.60	38.90	0.627	0.138
0.296	0.408	0.103	0.511	-0.0281	-0.0265	-0.0546	31.89	50.70	35.68	0.624	0.000
0.500	0.470	0.097	0.567	-0.0235	-0.0270	-0.0505	30.00	52.77	33.90	0.567	-0.099
0.697	0.519	0.084	0.603	-0.0295	-0.0258	-0.0552	30.67	55.58	34.15	0.475	-0.135
0.894	0.523	0.032	0.555	-0.0465	-0.0165	-0.0630	33.89	76.49	36.35	0.319	-0.104

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.541	0.541	0.541	0.541	0.541
0.01			0.412		
0.02			0.333		
0.03			0.282		
0.04			0.254		
0.05	0.229	0.213	0.222	0.197	0.111
0.10	0.153	0.147	0.123	0.104	0.037
0.15	0.114	0.098	0.077	0.046	-0.008
0.20	0.090	0.063	0.058	0.025	-0.034
0.30	0.063	0.029	0.013	-0.008	-0.086
0.40	0.032	0.018	0.002	-0.023	-0.078
0.50	0.014	-0.000	-0.006	-0.014	-0.068
0.55					
0.60	0.050	0.056	0.060	0.052	-0.004
0.65			0.109		
0.70	0.130	0.155	0.150	0.146	0.101
0.75	0.176	0.185	0.190	0.180	0.135
0.80	0.187	0.192	0.202	0.189	0.155
0.85	0.164	0.192	0.199	0.184	0.159
0.90	0.146	0.157	0.156	0.165	0.136
0.95	0.097	0.120	0.113	0.121	0.103
1.00	0.019	0.036	0.046	0.046	-0.011

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.425	0.340	0.245	0.188	0.218
0.003			-0.177		
0.006			-0.408	-0.439	
0.01	-0.296	-0.554	-0.669	-0.706	-0.607
0.02	-0.590	-0.921	-1.007	-0.984	-0.829
0.03	-0.673	-1.075	-1.176	-1.128	-0.910
0.04	-0.660	-1.145	-1.224	-1.150	-0.879
0.05	-0.650	-1.171	-1.295	-1.303	-1.044
0.06	-0.629	-1.110	-1.314	-1.281	-1.047
0.08	-0.598	-0.951	-1.276	-1.248	-1.015
0.10	-0.570	-0.869	-1.264	-1.240	-1.007
0.125	-0.542	-0.791	-1.235	-1.257	-0.987
0.15	-0.526	-0.707	-0.801	-1.255	-0.906
0.175	-0.511	-0.647	-0.722	-0.858	-0.751
0.20	-0.480	-0.603	-0.674	-0.711	-0.639
0.225	-0.467	-0.582	-0.645	-0.663	-0.654
0.25	-0.439	-0.555	-0.629	-0.657	-0.700
0.30	-0.431	-0.500	-0.575	-0.668	-0.849
0.35	-0.403	-0.482	-0.533	-0.652	-0.938
0.40	-0.385	-0.442	-0.500	-0.625	-0.925
0.45	-0.387	-0.411	-0.476	-0.591	-0.613
0.50	-0.357	-0.389	-0.430	-0.536	-0.444
0.55	-0.329	-0.355	-0.391	-0.454	-0.402
0.60	-0.301	-0.318	-0.350	-0.367	-0.362
0.65	-0.282	-0.296	-0.299	-0.307	-0.316
0.70	-0.261	-0.253	-0.251	-0.261	-0.273
0.75	-0.239	-0.221	-0.219	-0.222	-0.244
0.80	-0.198	-0.171	-0.173	-0.177	-0.209
0.85	-0.139	-0.124	-0.112	-0.121	-0.159
0.90	-0.072	-0.070	-0.053	-0.054	-0.117
0.95	-0.011	-0.006	0.008	-0.004	-0.068
1.00	0.019	0.036	0.046	0.046	-0.011

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR								
10.35	0.247	-0.003			-0.387				
11.35	0.270	0.076			-0.379				
12.35	0.294				-0.386		-0.326		
14.35	0.342	0.078			-0.393				
15.35	0.366	0.069			-0.387				-0.306
16.35	0.390	0.062			-0.360				-0.126
17.35	0.413	0.042			-0.364		-0.332		-0.288
18.35	0.437	0.038			-0.363				-0.294
19.35	0.461	0.041			-0.348				-0.275
20.35	0.485	0.038			-0.338				-0.273
22.35	0.533	0.037			-0.301				-0.249
23.35	0.556	0.044			-0.311				-0.255
24.35	0.580	0.067			-0.300		-0.284		-0.261
25.35	0.604	0.067			-0.293				-0.254
26.35	0.628	0.072			-0.280				-0.255
27.35	0.652	0.067			-0.280				
30.35	0.723								-0.213
31.35	0.747		0.131	-0.226		-0.215	-0.188	-0.223	-0.183
32.35	0.771		0.142	-0.214		-0.183		-0.203	-0.185
33.35	0.795		0.161	-0.181		-0.157		-0.183	-0.180
34.35	0.818		0.162	-0.149		-0.161		-0.166	-0.137
35.35	0.842		0.156	-0.125		-0.157		-0.144	-0.146
36.35	0.866		0.130	-0.119		-0.125		-0.116	-0.122
37.35	0.890		0.109	-0.093		-0.097		-0.118	-0.125
38.35	0.914		0.093	-0.062		-0.079		-0.099	-0.122
39.35	0.938		0.099	-0.034	-0.045	-0.052		-0.085	-0.097
40.35	0.961		0.082	0.009	-0.028	-0.047		-0.066	-0.095
41.35	0.985		0.072	0.046	0.008	-0.027		-0.051	-0.087
42.35	1.009		0.059	0.042	0.030	-0.009		-0.033	-0.131
44.85	1.069		0.046	0.027	0.044	-0.007		-0.012	-0.060
45.85	1.092		0.046	0.041	0.038				
46.85	1.116		0.046	0.022	0.021				

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR								
16.75	0.399			-0.261	-0.165	-0.120			
13.75	0.328		-0.244	-0.236	-0.183	-0.126			
10.75	0.256		-0.283	-0.261	-0.183	-0.122	-0.087	-0.131	
7.75	0.185	-0.326	-0.332	-0.284	-0.188	-0.126	-0.071		-0.025
6.75	0.161				-0.223	-0.116	-0.051	-0.033	-0.012
5.75	0.137	-0.363	-0.347	-0.295	-0.208	-0.108	-0.055	-0.020	-0.020
4.75	0.113				-0.212	-0.101	-0.028	-0.011	-0.005
4.25	0.101	-0.386	-0.364	-0.300					
3.75	0.089				-0.215	-0.125	-0.027	-0.009	-0.007
2.75	0.066				-0.221	-0.099	-0.026	0.020	0.006
1.75	0.042				-0.238	-0.122	-0.009	0.020	0.026
0.75	0.018				-0.226	-0.119	0.008	0.030	0.044
-0.25	-0.006						0.046	0.042	0.027
-1.25	-0.030					0.131	0.130	0.072	0.059	0.046
-2.25	-0.054					0.132	0.145	0.086	0.078	0.053
-2.75	-0.066			0.042	0.067				
-3.25	-0.077					0.116	0.136	0.081	0.075	0.049
-4.25	-0.101	0.084	0.053	0.063	0.109	0.110	0.078	0.078	0.057
-5.25	-0.125					0.105	0.103	0.081	0.076	0.066
-6.25	-0.149	0.090	0.069	0.067	0.111	0.101	0.072		-0.048
-9.25	-0.220		0.081	0.072	0.088	0.097	0.081		
-12.25	-0.292		0.076	0.066	0.079	0.079			
-15.25	-0.363			0.072	0.080	0.084			

RUN.SEQ
198.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.855	2.982	6.78	1493	926	543.2	473.8	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.467	0.082	0.548	-0.0308	-0.0128	-0.0436	0.2363	31.60	40.64	32.95	43.10	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.358	0.104	0.461	-0.0506	-0.0206	-0.0712	39.17	44.81	40.44	0.661	0.131
0.296	0.453	0.090	0.542	-0.0415	-0.0237	-0.0652	34.17	51.41	37.02	0.662	-0.011
0.500	0.539	0.085	0.624	-0.0389	-0.0281	-0.0670	32.21	58.03	35.73	0.624	-0.120
0.697	0.593	0.073	0.666	-0.0390	-0.0267	-0.0656	31.58	61.29	34.85	0.524	-0.152
0.894	0.587	0.011	0.598	-0.0628	-0.0153	-0.0782	35.71	165.4	38.08	0.343	-0.116

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.547	0.547	0.547	0.547	0.547
0.01			0.376		
0.02			0.306		
0.03			0.259		
0.04			0.223		
0.05	0.[illegible]5	0.209	0.189	0.163	0.065
0.10	0.[illegible]7	0.119	0.101	0.075	0.001
0.15	0.123	0.078	0.054	0.026	-0.045
0.20	0.092	0.051	0.030	0.003	-0.069
0.30	0.059	0.015	-0.006	-0.030	-0.121
0.40	0.031	-0.006	-0.027	-0.045	-0.124
0.50	0.005	-0.020	-0.037	-0.026	-0.104
0.55					
0.60	0.039	0.040	0.061	0.041	-0.011
0.65			0.114		
0.70	0.117	0.147	0.157	0.140	0.107
0.75	0.156	0.178	0.191	0.177	0.148
0.80	0.176	0.185	0.203	0.202	0.166
0.85	0.180	0.185	0.200	0.205	0.165
0.90	0.136	0.150	0.176	0.171	0.142
0.95	0.084	0.102	0.128	0.128	0.088
1.00	0.024	0.042	0.053	0.049	-0.067

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.486	0.410	0.328	0.286	0.296
0.003			-0.052		
0.006			-0.262	-0.289	
0.01	-0.205	-0.413	-0.502	-0.525	-0.443
0.02	-0.502	-0.758	-0.820	-0.793	-0.655
0.03	-0.601	-0.936	-0.981	-0.933	-0.740
0.04	-0.591	-0.999	-1.032	-0.958	-0.724
0.05	-0.586	-1.044	-1.107	-1.098	-0.881
0.06	-0.569	-1.027	-1.129	-1.092	-0.892
0.08	-0.544	-1.011	-1.120	-1.072	-0.875
0.10	-0.525	-0.859	-1.143	-1.087	-0.893
0.125	-0.503	-0.774	-1.141	-1.106	-0.909
0.15	-0.486	-0.762	-1.139	-1.135	-0.917
0.175	-0.482	-0.709	-1.116	-1.123	-0.931
0.20	-0.486	-0.668	-0.923	-1.125	-0.961
0.225	-0.478	-0.625	-0.776	-1.123	-0.980
0.25	-0.457	-0.600	-0.724	-1.114	-0.988
0.30	-0.457	-0.572	-0.677	-1.076	-1.054
0.35	-0.444	-0.551	-0.651	-0.775	-1.138
0.40	-0.447	-0.545	-0.652	-0.753	-0.759
0.45	-0.454	-0.531	-0.642	-0.757	-0.651
0.50	-0.448	-0.511	-0.626	-0.744	-0.627
0.55	-0.424	-0.467	-0.596	-0.649	-0.587
0.60	-0.402	-0.442	-0.556	-0.381	-0.527
0.65	-0.364	-0.397	-0.362	-0.265	-0.405
0.70	-0.330	-0.297	-0.244	-0.227	-0.325
0.75	-0.274	-0.254	-0.199	-0.201	-0.260
0.80	-0.219	-0.186	-0.170	-0.162	-0.209
0.85	-0.156	-0.128	-0.117	-0.111	-0.170
0.90	-0.082	-0.063	-0.054	-0.042	-0.119
0.95	-0.015	-0.007	0.008	0.014	-0.086
1.00	0.024	0.042	0.053	0.049	-0.067

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7 5	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR								
10.35	0.247	0.015			-0.384				
11.35	0.270	0.094			-0.391				
12.35	0.294				-0.400		-0.352		
14.35	0.342	0.064			-0.396				
15.35	0.366	0.058			-0.399				-0.349
16.35	0.390	0.052			-0.421				0.878
17.35	0.413	0.050			-0.419		-0.380		-0.341
18.35	0.437	0.044			-0.420				-0.356
19.35	0.461	0.043			-0.427				-0.340
20.35	0.485	0.039			-0.430				-0.336
22.35	0.533	0.036			-0.425				-0.345
23.35	0.556	0.035			-0.410				-0.346
24.35	0.580	0.044			-0.413		-0.395		-0.344
25.35	0.604	0.043			-0.404				-0.356
26.35	0.628	0.054			-0.375				-0.320
27.35	0.652	0.065			-0.351				
30.35	0.723								-0.281
31.35	0.747		0.131	-0.259		-0.286	-0.284	-0.277	-0.237
32.35	0.771		0.145	-0.235		-0.256		-0.254	-0.248
33.35	0.795		0.161	-0.230		-0.249		-0.234	-0.256
34.35	0.818		0.146	-0.195		-0.214		-0.204	-0.198
35.35	0.842		0.136	-0.162		-0.185		-0.176	-0.192
36.35	0.866		0.140	-0.138		-0.164		-0.147	-0.149
37.35	0.890		0.120	-0.118		-0.135		-0.137	-0.122
38.35	0.914		0.102	-0.076		-0.091		-0.105	-0.114
39.35	0.938		0.102	-0.026	-0.065	-0.063		-0.093	-0.103
40.35	0.961		0.085	-0.013	-0.038	-0.042		-0.087	-0.107
41.35	0.985		0.071	0.019	-0.010	-0.044		-0.074	-0.099
42.35	1.009		0.067	0.030	0.022	-0.020		-0.061	-0.151
44.85	1.069		0.044	0.034	0.035	-0.009		-0.038	-0.074
45.85	1.092		0.038	0.031	0.044				
46.85	1.116		0.031	0.027	0.016				

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR								
16.75	0.399			-0.346	-0.245	-0.182			
13.75	0.328		-0.296	-0.327	-0.275	-0.167			
10.75	0.256		-0.341	-0.344	-0.237	-0.149	-0.099	-0.151	
7.75	0.185	-0.352	-0.380	-0.395	-0.284	-0.155	-0.072		-0.052
6.75	0.161				-0.277	-0.147	-0.074	-0.061	-0.038
5.75	0.137	-0.388	-0.419	-0.423	-0.299	-0.142	-0.056	-0.012	-0.016
4.75	0.113				-0.288	-0.135	-0.056	-0.014	-0.035
4.25	0.101	-0.400	-0.419	-0.413					
3.75	0.089				-0.286	-0.164	-0.044	-0.020	-0.009
2.75	0.066				-0.329	-0.144	-0.035	-0.004	-0.006
1.75	0.042				-0.289	-0.137	-0.019	0.002	0.003
0.75	0.018				-0.259	-0.138	-0.010	0.022	0.035
-0.25	-0.006						0.019	0.030	0.034
-1.25	-0.030					0.131	0.140	0.071	0.067	0.044
-2.25	-0.054					0.147	0.129	0.073	0.069	0.056
-2.75	-0.066			0.050	0.044				
-3.25	-0.077					0.105	0.120	0.074	0.068	0.062
-4.25	-0.101	0.081	0.058	0.060		0.103	0.120	0.075	0.065	0.045
-5.25	-0.125					0.098	0.107	0.064	0.072	0.054
-6.25	-0.149	0.092	0.059	0.057		0.103	0.090	0.067		-0.037
-9.25	-0.220		0.061	0.054		0.084	0.074	0.071		
-12.25	-0.292		0.090	0.048		0.072	0.067			
-15.25	-0.363			0.051		0.078	0.058			

RUN.SEQ
199.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.852	2.990	6.80	1508	939	545.7	476.7	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.451	0.087	0.538	-0.0265	-0.0137	-0.0401	0.2332	30.87	40.72	32.46	43.34	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.340	0.107	0.447	-0.0449	-0.0217	-0.0666	38.20	45.30	39.90	0.641	0.132
0.296	0.442	0.089	0.531	-0.0379	-0.0235	-0.0614	33.58	51.41	36.56	0.648	-0.007
0.500	0.522	0.095	0.617	-0.0328	-0.0282	-0.0609	31.28	54.58	34.87	0.617	-0.113
0.697	0.560	0.086	0.646	-0.0295	-0.0281	-0.0576	30.27	57.63	33.92	0.508	-0.144
0.894	0.598	0.015	0.613	-0.0607	-0.0153	-0.0760	35.15	127.8	37.40	0.352	-0.117

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.547	0.547	0.547	0.547	0.547
0.01			0.391		
0.02			0.306		
0.03			0.258		
0.0[illegible]			0.233		
0.0[illegible]	0.240	0.205	0.201	0.189	0.090
0.10	0.171	0.129	0.106	0.093	0.005
0.15	0.136	0.081	0.078	0.036	-0.039
0.20	0.098	0.049	0.060	0.025	-0.063
0.30	0.060	0.017	0.024	-0.011	-0.119
0.40	0.031	-0.009	-0.011	-0.028	-0.114
0.50	0.008	-0.025	-0.010	-0.016	-0.095
0.55					
0.60	0.039	0.038	0.048	0.055	-0.013
0.65			0.104		
0.70	0.126	0.145	0.149	0.155	0.098
0.75	0.167	0.180	0.191	0.185	0.138
0.80	0.180	0.189	0.202	0.203	0.154
0.85	0.177	0.186	0.201	0.201	0.161
0.90	0.140	0.149	0.176	0.172	0.144
0.95	0.093	0.104	0.123	0.132	0.099
1.00	0.028	0.041	0.059	0.060	-0.030

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.483	0.408	0.319	0.276	0.292
0.003			-0.068		
0.006			-0.278	-0.305	
0.01	-0.216	-0.421	-0.519	-0.545	-0.458
0.02	-0.508	-0.769	-0.839	-0.816	-0.678
0.03	-0.606	-0.955	-1.000	-0.950	-0.764
0.04	-0.591	-1.012	-1.053	-0.976	-0.748
0.05	-0.590	-1.059	-1.135	-1.124	-0.900
0.06	-0.571	-1.034	-1.143	-1.113	-0.909
0.08	-0.549	-1.029	-1.137	-1.095	-0.891
0.10	-0.519	-0.866	-1.160	-1.108	-0.912
0.125	-0.499	-0.787	-1.156	-1.120	-0.924
0.15	-0.486	-0.764	-1.154	-1.149	-0.936
0.175	-0.475	-0.707	-1.117	-1.136	-0.948
0.20	-0.472	-0.664	-0.849	-1.135	-0.974
0.225	-0.453	-0.622	-0.749	-1.136	-0.998
0.25	-0.438	-0.590	-0.702	-1.117	-1.007
0.30	-0.449	-0.559	-0.663	-0.888	-1.071
0.35	-0.441	-0.534	-0.642	-0.718	-1.147
0.40	-0.438	-0.525	-0.642	-0.721	-0.927
0.45	-0.430	-0.506	-0.630	-0.709	-0.677
0.50	-0.417	-0.493	-0.603	-0.643	-0.637
0.55	-0.396	-0.450	-0.557	-0.450	-0.592
0.60	-0.350	-0.413	-0.406	-0.351	-0.505
0.65	-0.312	-0.334	-0.313	-0.269	-0.437
0.70	-0.287	-0.268	-0.258	-0.223	-0.343
0.75	-0.245	-0.231	-0.197	-0.183	-0.242
0.80	-0.192	-0.185	-0.161	-0.140	-0.168
0.85	-0.143	-0.136	-0.106	-0.091	-0.142
0.90	-0.086	-0.071	-0.047	-0.028	-0.113
0.95	-0.017	-0.011	0.017	0.023	-0.064
1.00	0.028	0.041	0.059	0.060	-0.030

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.025				-0.376				
11.35	0.270	0.090				-0.390				
12.35	0.294					-0.398		-0.340		
14.35	0.342	0.065				-0.401				
15.35	0.366	0.055				-0.401				-0.335
16.35	0.390	0.050				-0.402				0.729
17.35	0.413	0.044				-0.400		-0.368		-0.325
18.35	0.437	0.034				-0.394				-0.329
19.35	0.461	0.035				-0.395				-0.321
20.35	0.485	0.022				-0.395				-0.328
22.35	0.533	0.021				-0.383				-0.346
23.35	0.556	0.041				-0.386				-0.337
24.35	0.580	0.048				-0.386		-0.356		-0.326
25.35	0.604	0.061				-0.384				-0.316
26.35	0.628	0.071				-0.378				-0.325
27.35	0.652	0.073				-0.368				
30.35	0.723									-0.264
31.35	0.747		0.136		-0.255		-0.273	-0.238	-0.252	-0.236
32.35	0.771		0.147		-0.210		-0.246		-0.248	-0.213
33.35	0.795		0.144		-0.236		-0.222		-0.217	-0.184
34.35	0.818		0.141		-0.193		-0.166		-0.195	-0.200
35.35	0.842		0.139		-0.160		-0.156		-0.162	-0.147
36.35	0.866		0.133		-0.129		-0.147		-0.149	-0.143
37.35	0.890		0.123		-0.109		-0.120		-0.125	-0.127
38.35	0.914		0.105		-0.067		-0.085		-0.119	-0.117
39.35	0.938		0.099	-0.011	-0.027		-0.052		-0.094	-0.089
40.35	0.961		0.073	0.008	0.001		-0.042		-0.081	-0.075
41.35	0.985		0.054	0.034	0.023		-0.025		-0.056	-0.080
42.35	1.009		0.058	0.040	0.038		-0.018		-0.026	-0.150
44.85	1.069		0.041	0.023	0.031		-0.007		-0.022	-0.064
45.85	1.092		0.052	0.025	0.024					
46.85	1.116		0.037	0.028	0.029					

NORMAL PLWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.315	-0.241	-0.132			
13.75	0.328		-0.295	-0.301	-0.246	-0.149			
10.75	0.256		-0.325	-0.326	-0.236	-0.143	-0.080	-0.150	
7.75	0.185	-0.340	-0.368	-0.356	-0.238	-0.131	-0.066		-0.043
6.75	0.161				-0.252	-0.149	-0.056	-0.026	-0.022
5.75	0.137	-0.378	-0.384	-0.377	-0.245	-0.141	-0.064	-0.023	-0.028
4.75	0.113				-0.263	-0.133	-0.034	-0.023	-0.010
4.25	0.101	-0.398	-0.400	-0.386					
3.75	0.089				-0.273	-0.147	-0.025	-0.018	-0.007
2.75	0.066				-0.253	-0.151	-0.030	-0.002	0.019
1.75	0.042				-0.252	-0.147	-0.028	0.023	0.029
0.75	0.018				-0.255	-0.129	0.023	0.038	0.031
-0.25	-0.006						0.034	0.040	0.023
-1.25	-0.030				0.136	0.133	0.054	0.058	0.041
-2.25	-0.054				0.142	0.142	0.083	0.081	0.063
-2.75	-0.066		0.044	0.048					
-3.25	-0.077				0.123	0.132	0.088	0.079	0.066
-4.25	-0.101	0.082	0.056	0.041	0.110	0.130	0.081	0.070	0.057
-5.25	-0.125				0.104	0.119	0.091	0.082	0.047
-6.25	-0.149	0.090	0.067	0.046	0.107	0.119	0.088		-0.039
-9.25	-0.220		0.070	0.052	0.080	0.089	0.096		
-12.25	-0.292		0.062	0.055	0.067	0.078			
-15.25	-0.363			0.057	0.068	0.089			

RUN.SEQ
200.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.779	2.991	6.81	1578	1057	546.4	448.8	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.413	0.094	0.508	-0.0155	-0.0151	-0.0306	0.2169	28.75	41.02	31.02	42.74	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.333	0.100	0.433	-0.0381	-0.0196	-0.0576	36.41	44.61	38.30	0.621	0.142
0.296	0.411	0.101	0.512	-0.0266	-0.0250	-0.0517	31.48	49.78	35.09	0.625	0.005
0.500	0.465	0.107	0.571	-0.0234	-0.0290	-0.0525	30.04	52.26	34.18	0.571	-0.101
0.697	0.505	0.097	0.602	-0.0277	-0.0283	-0.0560	30.49	54.20	34.31	0.473	-0.135
0.894	0.506	0.042	0.548	-0.0467	-0.0181	-0.0648	34.24	68.08	36.83	0.314	-0.104

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.539	0.539	0.539	0.539	0.539
0.01			0.426		
0.02			0.349		
0.03			0.296		
0.04			0.255		
0.05	0.232	0.220	0.229	0.207	0.125
0.10	0.157	0.129	0.131	0.120	0.049
0.15	0.118	0.092	0.085	0.065	-0.000
0.20	0.091	0.065	0.068	0.043	-0.026
0.30	0.061	0.032	0.036	0.012	-0.065
0.40	0.032	0.019	0.002	0.001	-0.065
0.50	0.005	0.012	0.003	-0.010	-0.060
0.55					
0.60	0.034	0.054	0.066	0.055	0.011
0.65			0.117		
0.70	0.115	0.153	0.161	0.151	0.115
0.75	0.153	0.178	0.195	0.192	0.151
0.80	0.170	0.185	0.203	0.213	0.153
0.85	0.166	0.180	0.198	0.209	0.165
0.90	0.130	0.149	0.174	0.164	0.144
0.95	0.084	0.102	0.131	0.131	0.095
1.00	0.014	0.036	0.047	0.048	-0.015

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.414	0.319	0.211	0.160	0.202
0.003			-0.228		
0.006			-0.459	-0.500	
0.01	-0.332	-0.617	-0.732	-0.771	-0.639
0.02	-0.628	-0.977	-1.079	-1.058	-0.861
0.03	-0.718	-1.171	-1.260	-1.198	-0.922
0.04	-0.688	-1.202	-1.296	-1.216	-0.895
0.05	-0.680	-1.200	-1.375	-1.373	-1.110
0.06	-0.644	-1.154	-1.371	-1.361	-1.062
0.08	-0.610	-0.962	-1.319	-1.291	-0.782
0.10	-0.573	-0.858	-1.317	-1.283	-0.673
0.125	-0.552	-0.789	-0.833	-1.194	-0.740
0.15	-0.528	-0.724	-0.757	-0.762	-0.795
0.175	-0.505	-0.655	-0.708	-0.708	-0.809
0.20	-0.487	-0.608	-0.674	-0.707	-0.838
0.225	-0.469	-0.583	-0.646	-0.683	-0.858
0.25	-0.438	-0.552	-0.621	-0.679	-0.839
0.30	-0.434	-0.513	-0.591	-0.686	-0.883
0.35	-0.398	-0.448	-0.533	-0.667	-0.834
0.40	-0.387	-0.430	-0.502	-0.632	-0.661
0.45	-0.375	-0.407	-0.472	-0.559	-0.580
0.50	-0.343	-0.404	-0.438	-0.476	-0.501
0.55	-0.326	-0.357	-0.393	-0.427	-0.435
0.60	-0.317	-0.328	-0.344	-0.371	-0.377
0.65	-0.275	-0.291	-0.286	-0.313	-0.329
0.70	-0.257	-0.255	-0.251	-0.257	-0.272
0.75	-0.236	-0.222	-0.207	-0.225	-0.243
0.80	-0.191	-0.166	-0.170	-0.174	-0.199
0.85	-0.141	-0.114	-0.115	-0.119	-0.148
0.90	-0.098	-0.059	-0.061	-0.044	-0.106
0.95	-0.024	-0.008	0.005	0.009	-0.069
1.00	0.014	0.036	0.047	0.048	-0.015

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.021				-0.382				
11.35	0.270	0.087				-0.389				
12.35	0.294					-0.372		-0.333		
14.35	0.342	0.066				-0.366				
15.35	0.366	0.055				-0.357				-0.296
16.35	0.390	0.056				-0.362				0.442
17.35	0.413	0.045				-0.355		-0.329		-0.276
18.35	0.437	0.036				-0.358				-0.271
19.35	0.461	0.045				-0.343				-0.272
20.35	0.485	0.026				-0.334				-0.275
22.35	0.533	0.020				-0.304				-0.277
23.35	0.556	0.036				-0.318				-0.247
24.35	0.580	0.049				-0.303		-0.274		-0.256
25.35	0.604	0.044				-0.305				-0.258
26.35	0.628	0.050				-0.253				-0.233
27.35	0.652	0.072				-0.253				
30.35	0.723									-0.215
31.35	0.747		0.135		-0.235		-0.208	-0.198	-0.197	-0.169
32.35	0.771		0.140		-0.204		-0.194		-0.188	-0.160
33.35	0.795		0.146		-0.187		-0.186		-0.174	-0.157
34.35	0.818		0.139		-0.167		-0.162		-0.152	-0.130
35.35	0.842		0.143		-0.150		-0.130		-0.130	-0.132
36.35	0.866		0.131		-0.126		-0.111		-0.110	-0.111
37.35	0.890		0.117		-0.100		-0.091		-0.104	-0.101
38.35	0.914		0.092		-0.071		-0.089		-0.100	-0.106
39.35	0.938		0.082	-0.029	-0.042		-0.062		-0.085	-0.089
40.35	0.961		0.052	-0.017	-0.019		-0.046		-0.051	-0.067
41.35	0.985		0.039	0.017	0.003		-0.041		-0.038	-0.072
42.35	1.009		0.063	0.039	0.013		-0.021		-0.039	-0.131
44.85	1.069		0.037	0.028	0.007		-0.001		-0.019	-0.064
45.85	1.092		0.030	0.024	0.014					
46.85	1.116		0.041	0.014	0.023					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.264	-0.179	-0.115			
13.75	0.328		-0.235	-0.223	-0.193	-0.115			
10.75	0.256		-0.276	-0.256	-0.169	-0.111	-0.072	-0.131	
7.75	0.185	-0.333	-0.329	-0.274	-0.198	-0.115	-0.066		-0.037
6.75	0.161				-0.197	-0.110	-0.038	-0.039	-0.019
5.75	0.137	-0.360	-0.335	-0.294	-0.207	-0.118	-0.039	-0.017	-0.030
4.75	0.113				-0.204	-0.118	-0.047	-0.021	-0.003
4.25	0.101	-0.372	-0.355	-0.303					
3.75	0.089				-0.208	-0.111	-0.041	-0.021	-0.001
2.75	0.066				-0.213	-0.122	-0.017	0.006	0.006
1.75	0.042				-0.204	-0.128	0.008	0.002	0.021
0.75	0.018				-0.235	-0.126	0.003	0.013	0.007
-0.25	-0.006						0.017	0.039	0.028
-1.25	-0.030					0.135	0.131	0.039	0.063	0.037
-2.25	-0.054					0.116	0.128	0.078	0.074	0.037
-2.75	-0.066			0.045	0.049					
-3.25	-0.077					0.110	0.117	0.073	0.075	0.041
-4.25	-0.101	0.078	0.047	0.039	0.111	0.108	0.059	0.073	0.042
-5.25	-0.125					0.115	0.088	0.071	0.068	0.052
-6.25	-0.149	0.085	0.063	0.036	0.103	0.083	0.068		-0.062
-9.25	-0.220			0.071	0.047	0.078	0.071	0.075		
-12.25	-0.292			0.059	0.053	0.077	0.070			
-15.25	-0.363				0.063	0.068	0.068			

RUN.SEQ
201.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.741	2.993	6.81	1623	1127	546.6	433.1	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.395	0.098	0.493	-0.0139	-0.0147	-0.0286	0.2108	28.52	40.02	30.80	42.73	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.316	0.106	0.422	-0.0325	-0.0205	-0.0530	35.26	44.37	37.55	0.605	0.145
0.296	0.389	0.105	0.495	-0.0232	-0.0257	-0.0489	30.96	49.37	34.89	0.603	0.006
0.500	0.447	0.109	0.556	-0.0237	-0.0286	-0.0524	30.30	51.26	34.41	0.556	-0.099
0.697	0.486	0.100	0.586	-0.0284	-0.0272	-0.0556	30.84	52.15	34.49	0.461	-0.132
0.894	0.489	0.043	0.532	-0.0464	-0.0173	-0.0636	34.48	65.27	36.97	0.305	-0.101

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.535	0.535	0.535	0.535	0.535
0.01			0.432		
0.02			0.358		
0.03			0.313		
0.04			0.268		
0.05	0.236	0.234	0.235	0.216	0.141
0.10	0.165	0.136	0.131	0.128	0.053
0.15	0.129	0.100	0.094	0.071	0.008
0.20	0.100	0.070	0.073	0.057	-0.017
0.30	0.065	0.040	0.035	0.024	-0.074
0.40	0.037	0.022	0.013	0.011	-0.066
0.50	0.021	0.011	0.011	-0.002	-0.049
0.55					
0.60	0.048	0.055	0.080	0.059	0.010
0.65			0.124		
0.70	0.125	0.157	0.162	0.153	0.110
0.75	0.156	0.179	0.189	0.183	0.140
0.80	0.165	0.190	0.200	0.192	0.160
0.85	0.164	0.185	0.196	0.190	0.159
0.90	0.128	0.155	0.162	0.168	0.136
0.95	0.081	0.106	0.121	0.125	0.092
1.00	0.017	0.042	0.044	0.047	-0.006

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.378	0.277	0.162	0.104	0.178
0.003			-0.302		
0.006			-0.550	-0.589	
0.01	-0.381	-0.697	-0.836	-0.876	-0.678
0.02	-0.658	-1.071	-1.195	-1.176	-0.871
0.03	-0.732	-1.252	-1.370	-1.317	-0.918
0.04	-0.702	-1.182	-1.391	-1.314	-0.910
0.05	-0.678	-1.214	-1.460	-1.483	-1.036
0.06	-0.650	-1.009	-1.391	-1.403	-0.817
0.08	-0.611	-0.901	-0.980	-1.050	-0.830
0.10	-0.573	-0.825	-0.890	-0.862	-0.801
0.125	-0.538	-0.744	-0.843	-0.835	-0.788
0.15	-0.510	-0.681	-0.777	-0.827	-0.774
0.175	-0.480	-0.630	-0.719	-0.777	-0.769
0.20	-0.471	-0.586	-0.669	-0.742	-0.779
0.225	-0.451	-0.552	-0.630	-0.704	-0.756
0.25	-0.415	-0.510	-0.602	-0.684	-0.750
0.30	-0.412	-0.476	-0.562	-0.674	-0.780
0.35	-0.382	-0.432	-0.510	-0.639	-0.753
0.40	-0.363	-0.405	-0.480	-0.582	-0.669
0.45	-0.349	-0.385	-0.459	-0.538	-0.587
0.50	-0.325	-0.367	-0.418	-0.476	-0.494
0.55	-0.303	-0.331	-0.370	-0.416	-0.435
0.60	-0.277	-0.305	-0.339	-0.365	-0.378
0.65	-0.249	-0.257	-0.291	-0.312	-0.333
0.70	-0.224	-0.221	-0.255	-0.262	-0.280
0.75	-0.214	-0.197	-0.199	-0.210	-0.244
0.80	-0.176	-0.164	-0.168	-0.162	-0.195
0.85	-0.128	-0.123	-0.117	-0.112	-0.141
0.90	-0.078	-0.068	-0.061	-0.055	-0.104
0.95	-0.017	0.004	0.001	-0.004	-0.074
1.00	0.017	0.042	0.044	0.047	-0.006

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR								
10.35	0.247	0.007			-0.379				
11.35	0.270	0.092			-0.378				
12.35	0.294				-0.350		-0.309		
14.35	0.342	0.073			-0.343				
15.35	0.366	0.066			-0.336				-0.274
16.35	0.390	0.062			-0.339				C.268
17.35	0.413	0.054			-0.338		-0.305		-0.257
18.35	0.437	0.040			-0.323				-0.252
19.35	0.461	0.036			-0.317				-0.261
20.35	0.485	0.054			-0.308				-0.258
22.35	0.533	0.051			-0.286				-0.243
23.35	0.556	0.065			-0.290				-0.219
24.35	0.580	0.065			-0.273		-0.248		-0.213
25.35	0.604	0.061			-0.264				-0.215
26.35	0.628	0.072			-0.245				-0.217
27.35	0.652	0.082			-0.238				
30.35	0.723								-0.168
31.35	0.747		0.137	-0.180		-0.184	-0.190	-0.162	-0.166
32.35	0.771		0.150	-0.162		-0.169		-0.157	-0.155
33.35	0.795		0.153	-0.165		-0.168		-0.139	-0.151
34.35	0.818		0.150	-0.157		-0.148		-0.132	-0.137
35.35	0.842		0.149	-0.124		-0.129		-0.114	-0.127
36.35	0.866		0.131	-0.110		-0.110		-0.106	-0.108
37.35	0.890		0.121	-0.081		-0.091		-0.107	-0.109
38.35	0.914		0.098	-0.050		-0.067		-0.097	-0.094
39.35	0.938		0.102	-0.026	-0.024	-0.047		-0.083	-0.068
40.35	0.961		0.080	-0.003	-0.001	-0.048		-0.063	-0.080
41.35	0.985		0.067	0.038	0.024	-0.029		-0.038	-0.085
42.35	1.009		0.078	0.048	0.039	-0.012		-0.025	-0.133
44.85	1.069		0.050	0.038	0.032	0.006		-0.007	-0.042
45.85	1.092		0.047	0.043	0.036				
46.85	1.116		0.049	0.033	0.023				

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR							
16.75	0.399			-0.238	-0.155	-0.109		
13.75	0.328		-0.226	-0.199	-0.156	-0.110		
10.75	0.256		-0.257	-0.213	-0.166	-0.108	-0.085	-0.133
7.75	0.185	-0.309	-0.305	-0.248	-0.190	-0.103	-0 049	-0.040
6.75	0.161				-0.162	-0.106	-0.038	-0.025 -0.007
5.75	0.137	-0.328	-0.326	-0.263	-0.195	-0.110	-0.025	-0.016 -0.020
4.75	0.113				-0.192	-0.106	-0.025	-0.017 0.009
4.25	0.101	-0.350	-0.338	-0.273				
3.75	0.089				-0.184	-0.110	-0.029	-0.012 0.006
2.75	0.066				-0.191	-0.103	-0.002	-0.005 0.018
1.75	0.042				-0.209	-0.100	0.004	0.027 0.030
0.75	0.018				-0.180	-0.110	0.024	0.039 0.032
-0.25	-0.006						0.038	0.048 0.038
-1.25	-0.030				0.137	0.131	0.067	0.078 0.050
-2.25	-0.054				0.131	0.131	0.089	0.079 0.063
-2.75	-0.066		0.054	0.065				
-3.25	-0.077				0.120	0.125	0.085	0.071 0.059
-4.25	-0.101	0.088	0.062	0.053	0.116	0.120	0.075	0.082 0.070
-5.25	-0.125				0.114	0.120	0.076	0.079 0.055
-6.25	-0.149	0.096	0.072	0.064	0.101	0.105	0.082	-0.048
-9.25	-0.220		0.081	0.083	0.079	0.115	0.083	
-12.25	-0.292		0.064	0.083	0.077	0.101		
-15.25	-0.363			0.081	0.076	0.083		

RUN.SEQ
202.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.700	2.997	6.82	1678	1209	546.3	415.1	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.382	0.101	0.483	-0.0120	-0.0150	-0.0270	0.2060	28.13	39.83	30.59	42.64	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.305	0.110	0.415	-0.0287	-0.0223	-0.0510	34.41	45.24	37.28	0.595	0.145
0.296	0.380	0.107	0.487	-0.0222	-0.0255	-0.0477	30.85	48.79	34.79	0.595	0.007
0.500	0.431	0.111	0.542	-0.0210	-0.0274	-0.0484	29.86	49.67	33.92	0.542	-0.094
0.697	0.469	0.105	0.574	-0.0291	-0.0275	-0.0566	31.19	51.28	34.86	0.452	-0.131
0.894	0.464	0.051	0.514	-0.0441	-0.0187	-0.0628	34.52	61.73	37.21	0.295	-0.098

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.531	0.531	0.531	0.531	0.531
0.01			0.445		
0.02			0.368		
0.03			0.323		
0.04			0.278		
0.05	0.237	0.233	0.248	0.231	0.145
0.10	0.159	0.149	0.146	0.145	0.059
0.15	0.122	0.104	0.104	0.090	0.013
0.20	0.098	0.073	0.080	0.059	-0.010
0.30	0.069	0.047	0.042	0.024	-0.047
0.40	0.043	0.026	0.019	0.010	-0.055
0.50	0.030	0.014	0.015	0.004	-0.046
0.55					
0.60	0.060	0.057	0.069	0.066	0.021
0.65			0.113		
0.70	0.131	0.151	0.154	0.155	0.117
0.75	0.156	0.178	0.189	0.190	0.151
0.80	0.168	0.191	0.195	0.197	0.160
0.85	0.164	0.183	0.191	0.194	0.158
0.90	0.134	0.149	0.163	0.163	0.140
0.95	0.090	0.111	0.116	0.124	0.096
1.00	0.019	0.030	0.047	0.044	-0.009

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.362	0.235	0.110	0.053	0.154
0.003			-0.371		
0.006			-0.632	-0.655	
0.01	-0.426	-0.773	-0.936	-0.955	-0.705
0.02	-0.700	-1.139	-1.306	-1.254	-0.866
0.03	-0.749	-1.237	-1.427	-1.317	-0.891
0.04	-0.713	-1.172	-1.333	-1.223	-0.853
0.05	-0.683	-1.114	-1.395	-1.445	-0.944
0.06	-0.657	-0.955	-1.118	-1.026	-0.772
0.08	-0.612	-0.878	-0.970	-0.955	-0.773
0.10	-0.574	-0.796	-0.923	-0.900	-0.752
0.125	-0.525	-0.730	-0.815	-0.845	-0.743
0.15	-0.505	-0.675	-0.757	-0.819	-0.726
0.175	-0.475	-0.615	-0.696	-0.752	-0.723
0.20	-0.456	-0.573	-0.665	-0.724	-0.725
0.225	-0.431	-0.532	-0.626	-0.693	-0.713
0.25	-0.401	-0.502	-0.601	-0.676	-0.709
0.30	-0.395	-0.466	-0.540	-0.649	-0.731
0.35	-0.360	-0.423	-0.495	-0.597	-0.693
0.40	-0.345	-0.395	-0.461	-0.547	-0.630
0.45	-0.327	-0.366	-0.424	-0.509	-0.561
0.50	-0.308	-0.339	-0.384	-0.461	-0.490
0.55	-0.288	-0.312	-0.348	-0.407	-0.429
0.60	-0.259	-0.291	-0.320	-0.362	-0.367
0.65	-0.230	-0.250	-0.274	-0.304	-0.321
0.70	-0.211	-0.217	-0.237	-0.256	-0.264
0.75	-0.195	-0.197	-0.190	-0.218	-0.227
0.80	-0.158	-0.163	-0.160	-0.172	-0.186
0.85	-0.120	-0.118	-0.108	-0.116	-0.137
0.90	-0.075	-0.063	-0.058	-0.060	-0.096
0.95	-0.011	-0.006	0.002	-0.005	-0.062
1.00	0.019	0.030	0.047	0.044	-0.009

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.019				-0.354				
11.35	0.270	0.085				-0.351				
12.35	0.294					-0.357		-0.288		
14.35	0.342	0.078				-0.348				
15.35	0.366	0.073				-0.321				-0.266
16.35	0.390	0.066				-0.326				0.081
17.35	0.413	0.056				-0.310		-0.275		-0.250
18.35	0.437	0.039				-0.314				-0.251
19.35	0.461	0.047				-0.315				-0.227
20.35	0.485	0.050				-0.299				-0.243
22.35	0.533	0.053				-0.272				-0.225
23.35	0.556	0.057				-0.268				-0.215
24.35	0.580	0.069				-0.271		-0.233		-0.215
25.35	0.604	0.060				-0.252				-0.200
26.35	0.628	0.070				-0.245				-0.190
27.35	0.652	0.089				-0.237				
30.35	0.723									-0.174
31.35	0.747		0.138		-0.186		-0.184	-0.158	-0.168	-0.164
32.35	0.771		0.147		-0.167		-0.168		-0.152	-0.139
33.35	0.795		0.147		-0.159		-0.164		-0.132	-0.132
34.35	0.818		0.139		-0.146		-0.141		-0.130	-0.130
35.35	0.842		0.138		-0.123		-0.128		-0.119	-0.113
36.35	0.866		0.132		-0.106		-0.116		-0.104	-0.121
37.35	0.890		0.116		-0.086		-0.098		-0.095	-0.103
38.35	0.914		0.097		-0.049		-0.066		-0.081	-0.090
39.35	0.938		0.097	-0.018	-0.030		-0.064		-0.064	-0.077
40.35	0.961		0.072	0.007	-0.014		-0.037		-0.056	-0.066
41.35	0.985		0.052	0.039	0.010		-0.018		-0.051	-0.068
42.35	1.009		0.064	0.050	0.023		-0.011		-0.035	-0.123
44.85	1.069		0.038	0.022	0.018		0.001		-0.026	-0.046
45.85	1.092		0.040	0.025	0.023					
46.85	1.116		0.045	0.033	0.027					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.213	-0.145	-0.112			
13.75	0.328		-0.211	-0.191	-0.143	-0.112			
10.75	0.256		-0.250	-0.215	-0.164	-0.121	-0.068	-0.123	
7.75	0.185	-0.288	-0.275	-0.233	-0.158	-0.106	-0.038		-0.031
6.75	0.161				-0.168	-0.104	-0.051	-0.035	-0.026
5.75	0.137	-0.318	-0.299	-0.256	-0.170	-0.105	-0.026	-0.017	-0.009
4.75	0.113				-0.165	-0.101	-0.022	-0.013	0.005
4.25	0.101	-0.357	-0.310	-0.271					
3.75	0.089				-0.184	-0.116	-0.018	-0.011	0.001
2.75	0.066				-0.174	-0.094	-0.012	0.003	0.018
1.75	0.042				-0.198	-0.104	-0.009	0.023	0.030
0.75	0.018				-0.186	-0.106	0.010	0.023	0.018
-0.25	-0.006						0.039	0.050	0.022
-1.25	-0.030				0.138	0.132	0.052	0.064	0.038
-2.25	-0.054				0.128	0.116	0.072	0.071	0.053
-2.75	-0.066		0.056	0.069					
-3.25	-0.077				0.124	0.117	0.070	0.068	0.050
-4.25	-0.101	0.091	0.063	0.054	0.106	0.114	0.079	0.077	0.046
-5.25	-0.125				0.097	0.098	0.078	0.071	0.057
-6.25	-0.149	0.096	0.072	0.067	0.102	0.093	0.082		-0.043
-9.25	-0.220		0.078	0.076	0.076	0.086	0.074		
-12.25	-0.292		0.076	0.072	0.083	0.074			
-15.25	-0.363			0.070	0.068	0.073			

RUN.SEQ
203.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.599	3.037	6.91	1878	1474	544.7	369.7	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.365	0.101	0.466	-0.0100	-0.0145	-0.0245	0.1986	27.74	39.37	30.26	42.62	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.234	0.105	0.399	-0.0267	-0.0201	-0.0468	34.08	44.19	36.73	0.571	0.144
0.296	0.367	0.106	0.473	-0.0201	-0.0240	-0.0441	30.47	47.64	34.32	0.577	0.010
0.500	0.412	0.113	0.525	-0.0200	-0.0268	-0.0468	29.85	48.71	33.91	0.525	-0.091
0.697	0.443	0.108	0.552	-0.0269	-0.0268	-0.0537	31.06	49.81	34.74	0.434	-0.126
0.894	0.434	0.058	0.492	-0.0426	-0.0184	-0.0610	34.82	56.66	37.41	0.282	-0.094

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.523	0.523	0.523	0.523	0.523
0.01			0.454		
0.02			0.384		
0.03			0.328		
0.04			0.291		
0.05	0.230	0.239	0.249	0.244	0.160
0.10	0.161	0.149	0.153	0.153	0.081
0.15	0.123	0.104	0.105	0.098	0.033
0.20	0.093	0.077	0.087	0.066	0.010
0.30	0.066	0.043	0.053	0.034	-0.034
0.40	0.041	0.036	0.026	0.021	-0.043
0.50	0.031	0.014	0.009	0.012	-0.038
0.55					
0.60	0.057	0.063	0.078	0.067	0.026
0.65			0.120		
0.70	0.121	0.143	0.157	0.155	0.117
0.75	0.145	0.170	0.181	0.182	0.147
0.80	0.154	0.180	0.189	0.196	0.152
0.85	0.147	0.176	0.188	0.190	0.156
0.90	0.129	0.147	0.158	0.160	0.138
0.95	0.081	0.093	0.112	0.117	0.096
1.00	0.009	0.026	0.036	0.035	-0.005

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.284	0.137	0.001	-0.039	0.109
0.003			-0.516		
0.006			-0.782	-0.779	
0.01	-0.502	-0.904	-1.068	-1.060	-0.717
0.02	-0.752	-1.205	-1.332	-1.242	-0.830
0.03	-0.762	-1.202	-1.326	-1.213	-0.829
0.04	-0.717	-1.127	-1.241	-1.146	-0.801
0.05	-0.694	-1.039	-1.175	-1.171	-0.829
0.06	-0.648	-0.928	-1.044	-0.993	-0.729
0.08	-0.593	-0.854	-0.935	-0.910	-0.720
0.10	-0.549	-0.784	-0.874	-0.862	-0.697
0.125	-0.525	-0.708	-0.776	-0.795	-0.673
0.15	-0.497	-0.642	-0.714	-0.775	-0.675
0.175	-0.472	-0.585	-0.664	-0.720	-0.663
0.20	-0.432	-0.544	-0.636	-0.682	-0.664
0.225	-0.408	-0.514	-0.593	-0.648	-0.658
0.25	-0.377	-0.482	-0.566	-0.626	-0.655
0.30	-0.362	-0.445	-0.519	-0.605	-0.667
0.35	-0.331	-0.399	-0.471	-0.560	-0.627
0.40	-0.318	-0.370	-0.440	-0.507	-0.580
0.45	-0.305	-0.347	-0.405	-0.476	-0.526
0.50	-0.285	-0.323	-0.374	-0.428	-0.463
0.55	-0.267	-0.295	-0.331	-0.385	-0.407
0.60	-0.244	-0.271	-0.288	-0.330	-0.352
0.65	-0.221	-0.239	-0.249	-0.284	-0.317
0.70	-0.205	-0.208	-0.226	-0.241	-0.255
0.75	-0.186	-0.185	-0.183	-0.205	-0.217
0.80	-0.152	-0.157	-0.159	-0.170	-0.186
0.85	-0.119	-0.113	-0.110	-0.117	-0.136
0.90	-0.073	-0.067	-0.059	-0.056	-0.096
0.95	-0.021	-0.003	-0.002	-0.003	-0.057
1.00	0.009	0.026	0.036	0.035	-0.005

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.047				-0.329				
11.35	0.270	0.089				-0.324				
12.35	0.294					-0.323		-0.267		
14.35	0.342	0.078				-0.317				
15.35	0.366	0.079				-0.313				-0.241
16.35	0.390	0.082				-0.296				-0.511
17.35	0.413	0.070				-0.292		-0.260		-0.223
18.35	0.437	0.057				-0.284				-0.227
19.35	0.461	0.057				-0.283				-0.206
20.35	0.485	0.055				-0.280				-0.214
22.35	0.533	0.060				-0.251				-0.205
23.35	0.556	0.065				-0.243				-0.195
24.35	0.580	0.082				-0.231		-0.214		-0.201
25.35	0.604	0.068				-0.231				-0.190
26.35	0.628	0.083				-0.210				-0.189
27.35	0.652	0.091				-0.206				
30.35	0.723									-0.152
31.35	0.747		0.138		-0.183		-0.168	-0.143	-0.152	-0.146
32.35	0.771		0.151		-0.162		-0.153		-0.156	-0.132
33.35	0.795		0.147		-0.146		-0.144		-0.142	-0.128
34.35	0.818		0.140		-0.128		-0.122		-0.130	-0.124
35.35	0.842		0.137		-0.108		-0.118		-0.121	-0.119
36.35	0.866		0.129		-0.090		-0.111		-0.108	-0.104
37.35	0.890		0.119		-0.069		-0.090		-0.081	-0.086
38.35	0.914		0.105		-0.051		-0.062		-0.074	-0.090
39.35	0.938		0.094	-0.017	-0.027		-0.050		-0.064	-0.066
40.35	0.961		0.071	0.000	-0.013		-0.026		-0.051	-0.065
41.35	0.985		0.073	0.036	0.017		-0.022		-0.044	-0.070
42.35	1.009		0.065	0.052	0.024		-0.008		-0.038	-0.113
44.85	1.069		0.044	0.027	0.023		0.001		-0.013	-0.051
45.85	1.092		0.050	0.028	0.020					
46.85	1.116		0.037	0.033	0.031					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.189	-0.127	-0.097			
13.75	0.328		-0.197	-0.169	-0.140	-0.098			
10.75	0.256		-0.223	-0.201	-0.146	-0.104	-0.070	-0.113	
7.75	0.185	-0.267	-0.260	-0.214	-0.143	-0.099	-0.060		-0.039
6.75	0.161				-0.152	-0.108	-0.044	-0.038	-0.013
5.75	0.137	-0.304	-0.282	-0.225	-0.158	-0.099	-0.028	-0.024	-0.033
4.75	0.113				-0.161	-0.101	-0.035	-0.023	-0.019
4.25	0.101	-0.323	-0.292	-0.231					
3.75	0.089				-0.168	-0.111	-0.022	-0.008	0.001
2.75	0.066				-0.178	-0.093	-0.017	0.002	-0.008
1.75	0.042				-0.170	-0.095	-0.011	0.011	0.024
0.75	0.018				-0.183	-0.090	0.017	0.024	0.023
-0.25	-0.006						0.036	0.052	0.027
-1.25	-0.030					0.138	0.129	0.073	0.065	0.044
-2.25	-0.054					0.135	0.123	0.092	0.077	0.055
-2.75	-0.066			0.070	0.082					
-3.25	-0.077					0.116	0.115	0.077	0.074	0.051
-4.25	-0.101	0.084	0.077	0.068		0.110	0.119	0.071	0.082	0.059
-5.25	-0.125					0.109	0.103	0.070	0.074	0.041
-6.25	-0.149	0.096	0.073	0.074		0.108	0.100	0.076		-0.022
-9.25	-0.220		0.086	0.070		0.085	0.083	0.081		
-12.25	-0.292		0.085	0.071		0.090	0.082			
-15.25	-0.363			0.077		0.077	0.074			

RUN.SEQ
204.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.501	3.023	6.88	2133	1798	543.9	315.2	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.350	0.106	0.456	-0.0074	-0.0150	-0.0225	0.1940	27.12	39.20	29.93	42.55	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.283	0.110	0.393	-0.0229	-0.0225	-0.0453	33.07	45.49	36.54	0.763	0.143
0.296	0.348	0.113	0.461	-0.0153	-0.0250	-0.0403	29.41	47.11	33.76	0.562	0.013
0.500	0.394	0.118	0.512	-0.0177	-0.0268	-0.0446	29.49	47.78	33.70	0.512	-0.088
0.697	0.428	0.111	0.539	-0.0259	-0.0258	-0.0517	31.06	48.25	34.60	0.424	-0.122
0.894	0.420	0.062	0.482	-0.0418	-0.0182	-0.0601	34.96	54.56	37.47	0.277	-0.092

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.516	0.516	0.516	0.516	0.516
0.01			0.462		
0.02			0.397		
0.03			0.336		
0.04			0.300		
0.05	0.224	0.252	0.257	0.251	0.167
0.10	0.161	0.156	0.164	0.149	0.087
0.15	0.119	0.103	0.116	0.103	0.043
0.20	0.095	0.089	0.098	0.081	0.015
0.30	0.062	0.055	0.056	0.052	-0.028
0.40	0.052	0.047	0.035	0.029	-0.034
0.50	0.043	0.028	0.019	0.022	-0.023
0.55					
0.60	0.072	0.070	0.087	0.061	0.029
0.65			0.128		
0.70	0.126	0.138	0.154	0.153	0.118
0.75	0.150	0.171	0.179	0.179	0.140
0.80	0.149	0.182	0.175	0.196	0.151
0.85	0.153	0.180	0.184	0.182	0.146
0.90	0.132	0.143	0.153	0.153	0.135
0.95	0.096	0.097	0.121	0.108	0.098
1.00	0.023	0.041	0.037	0.022	-0.016

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.286	0.064	-0.083	-0.112	0.073
0.003			-0.593		
0.006			-0.860	-0.841	
0.01	-0.514	-0.972	-1.124	-1.089	-0.725
0.02	-0.747	-1.193	-1.299	-1.210	-0.819
0.03	-0.768	-1.170	-1.266	-1.168	-0.811
0.04	-0.728	-1.082	-1.193	-1.093	-0.789
0.05	-0.694	-0.997	-1.120	-1.080	-0.768
0.06	-0.649	-0.907	-1.003	-0.971	-0.708
0.08	-0.593	-0.825	-0.895	-0.885	-0.693
0.10	-0.553	-0.755	-0.829	-0.829	-0.670
0.125	-0.510	-0.684	-0.760	-0.766	-0.656
0.15	-0.490	-0.617	-0.702	-0.745	-0.657
0.175	-0.449	-0.565	-0.641	-0.686	-0.645
0.20	-0.420	-0.528	-0.602	-0.660	-0.639
0.225	-0.393	-0.495	-0.560	-0.634	-0.625
0.25	-0.367	-0.470	-0.539	-0.608	-0.616
0.30	-0.361	-0.420	-0.499	-0.571	-0.633
0.35	-0.329	-0.381	-0.449	-0.528	-0.605
0.40	-0.311	-0.344	-0.413	-0.484	-0.562
0.45	-0.291	-0.325	-0.380	-0.449	-0.504
0.50	-0.267	-0.302	-0.341	-0.415	-0.446
0.55	-0.249	-0.278	-0.314	-0.367	-0.388
0.60	-0.232	-0.245	-0.277	-0.328	-0.345
0.65	-0.215	-0.212	-0.249	-0.274	-0.316
0.70	-0.187	-0.178	-0.210	-0.239	-0.261
0.75	-0.164	-0.163	-0.171	-0.197	-0.213
0.80	-0.130	-0.137	-0.140	-0.168	-0.179
0.85	-0.103	-0.104	-0.101	-0.110	-0.127
0.90	-0.061	-0.048	-0.053	-0.057	-0.095
0.95	-0.015	0.003	-0.006	-0.002	-0.061
1.00	0.023	0.041	0.037	0.022	-0.016

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	0.068				-0.328				
11.35	0.270	0.072				-0.307				
12.35	0.294					-0.318		-0.252		
14.35	0.342	0.073				-0.299				
15.35	0.366	0.066				-0.287				-0.217
16.35	0.390	0.076				-0.269				-1.253
17.35	0.413	0.075				-0.281		-0.247		-0.213
18.35	0.437	0.044				-0.258				-0.197
19.35	0.461	0.068				-0.263				-0.196
20.35	0.485	0.044				-0.257				-0.203
22.35	0.533	0.051				-0.228				-0.185
23.35	0.556	0.070				-0.222				-0.178
24.35	0.580	0.083				-0.233		-0.207		-0.180
25.35	0.604	0.057				-0.205				-0.170
26.35	0.628	0.096				-0.206				-0.159
27.35	0.652	0.084				-0.201				
30.35	0.723									-0.127
31.35	0.747		0.131		-0.170		-0.146	-0.132	-0.123	-0.136
32.35	0.771		0.143		-0.134		-0.125		-0.140	-0.133
33.35	0.795		0.155		-0.121		-0.150		-0.108	-0.114
34.35	0.818		0.125		-0.131		-0.101		-0.122	-0.099
35.35	0.842		0.142		-0.086		-0.102		-0.107	-0.118
36.35	0.866		0.125		-0.092		-0.096		-0.081	-0.077
37.35	0.890		0.111		-0.070		-0.066		-0.068	-0.083
38.35	0.914		0.107		-0.034		-0.041		-0.085	-0.084
39.35	0.938		0.105	-0.020	-0.010		-0.056		-0.047	-0.054
40.35	0.961		0.065	-0.001	-0.014		-0.021		-0.054	-0.055
41.35	0.985		0.081	0.041	0.034		-0.020		-0.045	-0.064
42.35	1.009		0.058	0.067	0.025		-0.008		-0.023	-0.076
44.85	1.069		0.036	0.029	0.031		0.018		-0.008	-0.050
45.85	1.092		0.049	0.046	0.037					
46.85	1.116		0.055	0.024	0.042					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.147	-0.125	-0.086			
13.75	0.328		-0.166	-0.149	-0.111	-0.099			
10.75	0.256		-0.213	-0.180	-0.136	-0.077	-0.064	-0.076	
7.75	0.185	-0.252	-0.247	-0.207	-0.132	-0.081	-0.059		-0.027
6.75	0.161				-0.123	-0.081	-0.045	-0.023	-0.008
5.75	0.137	-0.288	-0.252	-0.221	-0.150	-0.074	-0.018	-0.010	-0.015
4.75	0.113				-0.151	-0.090	-0.016	-0.015	-0.000
4.25	0.101	-0.318	-0.281	-0.233					
3.75	0.089				-0.146	-0.096	-0.020	-0.008	0.018
2.75	0.066				-0.147	-0.087	-0.013	0.007	-0.011
1.75	0.042				-0.170	-0.076	-0.007	0.031	0.036
0.75	0.018				-0.170	-0.092	0.034	0.025	0.031
-0.25	-0.006						0.041	0.067	0.029
-1.25	-0.030				0.131	0.125	0.081	0.058	0.036
-2.25	-0.054				0.133	0.111	0.093	0.095	0.049
-2.75	-0.066			0.075	0.083				
-3.25	-0.077				0.109	0.120	0.061	0.066	0.059
-4.25	-0.101	0.069	0.053	0.052	0.108	0.120	0.081	0.089	0.067
-5.25	-0.125				0.112	0.083	0.075	0.068	0.046
-6.25	-0.149	0.090	0.060	0.077	0.114	0.105	0.072		0.028
-9.25	-0.220		0.085	0.072	0.074	0.081	0.086		
-12.25	-0.292		0.081	0.069	0.087	0.073			
-15.25	-0.363			0.077	0.068	0.079			

RUN.SEQ
205.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.820	2.494	5.67	1277	821	544.8	386.6	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.435	0.084	0.519	-0.0191	-0.0133	-0.0324	0.2220	29.39	40.80	31.25	42.74	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.348	0.098	0.447	-0.0431	-0.0193	-0.0624	37.37	44.63	38.97	0.640	0.141
0.296	0.426	0.093	0.519	-0.0298	-0.0243	-0.0541	32.00	50.96	35.42	0.633	0.002
0.500	0.488	0.091	0.579	-0.0263	-0.0271	-0.0534	30.39	54.84	34.23	0.579	-0.102
0.697	0.543	0.083	0.626	-0.0293	-0.0268	-0.0561	30.40	57.41	33.96	0.493	-0.139
0.894	0.538	0.019	0.556	-0.0459	-0.0155	-0.0614	33.54	108.4	36.03	0.319	-0.104

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.543	0.543	0.543	0.543	0.543
0.01			0.399		
0.02			0.323		
0.03			0.276		
0.04			0.234		
0.05	0.225	0.222	0.202	0.182	0.112
0.10	0.151	0.124	0.105	0.089	0.005
0.15	0.115	0.079	0.065	0.036	-0.044
0.20	0.088	0.055	0.045	0.017	-0.066
0.30	0.058	0.019	0.009	-0.008	-0.112
0.40	0.030	0.001	-0.013	-0.022	-0.105
0.50	0.007	-0.012	-0.014	-0.020	-0.085
0.55					
0.60	0.040	0.042	0.055	0.055	-0.011
0.65			0.104		
0.70	0.124	0.144	0.145	0.152	0.097
0.75	0.150	0.168	0.195	0.176	0.136
0.80	0.163	0.190	0.208	0.194	0.167
0.85	0.151	0.191	0.197	0.197	0.167
0.90	0.128	0.158	0.157	0.165	0.139
0.95	0.080	0.108	0.112	0.123	0.078
1.00	0.011	0.043	0.059	0.044	-0.003

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.422	0.364	0.268	0.220	0.239
0.003			-0.144		
0.006			-0.362	-0.392	
0.01	-0.281	-0.505	-0.618	-0.644	-0.559
0.02	-0.570	-0.866	-0.947	-0.924	-0.782
0.03	-0.668	-1.056	-1.109	-1.067	-0.865
0.04	-0.653	-1.107	-1.163	-1.097	-0.844
0.05	-0.638	-1.145	-1.249	-1.230	-0.988
0.06	-0.622	-1.123	-1.259	-1.217	-0.995
0.08	-0.597	-1.081	-1.241	-1.204	-0.994
0.10	-0.570	-0.862	-1.250	-1.216	-1.000
0.125	-0.530	-0.826	-1.230	-1.228	-1.002
0.15	-0.511	-0.770	-1.069	-1.246	-0.991
0.175	-0.508	-0.711	-0.777	-1.218	-0.995
0.20	-0.505	-0.650	-0.720	-1.207	-1.004
0.225	-0.493	-0.605	-0.673	-0.939	-0.915
0.25	-0.465	-0.565	-0.655	-0.706	-0.864
0.30	-0.459	-0.535	-0.621	-0.690	-0.809
0.35	-0.440	-0.504	-0.571	-0.665	-0.773
0.40	-0.434	-0.488	-0.526	-0.643	-0.752
0.45	-0.403	-0.454	-0.502	-0.583	-0.683
0.50	-0.395	-0.405	-0.464	-0.533	-0.472
0.55	-0.379	-0.371	-0.415	-0.455	-0.425
0.60	-0.335	-0.345	-0.362	-0.378	-0.370
0.65	-0.299	-0.300	-0.311	-0.319	-0.326
0.70	-0.274	-0.247	-0.271	-0.266	-0.278
0.75	-0.241	-0.217	-0.223	-0.216	-0.247
0.80	-0.206	-0.187	-0.185	-0.169	-0.209
0.85	-0.153	-0.128	-0.121	-0.116	-0.168
0.90	-0.085	-0.070	-0.053	-0.057	-0.124
0.95	-0.032	-0.004	0.017	0.000	-0.064
1.00	0.011	0.043	0.059	0.044	-0.003

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID		1A	1B	2	3	4A	4B	5A	5B	6
Y		-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR		-.066	-.030	-.006	.018	.101	.089	.185	.161	.256
X	X/CR									
10.35	0.247	-0.019				-0.380				
11.35	0.270	0.092				-0.395				
12.35	0.294					-0.397		-0.362		
14.35	0.342	0.080				-0.396				
15.35	0.366	0.070				-0.405				-0.324
16.35	0.390	0.061				-0.411				1.304
17.35	0.413	0.055				-0.384		-0.353		-0.319
18.35	0.437	0.041				-0.386				-0.326
19.35	0.461	0.048				-0.356				-0.321
20.35	0.485	0.033				-0.363				-0.320
22.35	0.533	0.032				-0.332				-0.326
23.35	0.556	0.035				-0.372				-0.297
24.35	0.580	0.041				-0.344		-0.312		-0.299
25.35	0.604	0.044				-0.313				-0.281
26.35	0.628	0.070				-0.301				-0.255
27.35	0.652	0.086				-0.306				
30.35	0.723									-0.218
31.35	0.747		0.140		-0.249		-0.234	-0.234	-0.243	-0.219
32.35	0.771		0.151		-0.233		-0.220		-0.233	-0.203
33.35	0.795		0.139		-0.202		-0.204		-0.207	-0.180
34.35	0.818		.144		-0.161		-0.165		-0.188	-0.174
35.35	0.842		.138		-0.141		-0.156		-0.171	-0.149
36.35	0.866		0.140		-0.133		-0.142		-0.145	-0.143
37.35	0.890		0.120		-0.102		-0.123		-0.118	-0.143
38.35	0.914		0.095		-0.084		-0.115		-0.103	-0.119
39.35	0.938		0.078	-0.051	-0.056		-0.081		-0.082	-0.110
40.35	0.961		0.057	-0.021	-0.019		-0.051		-0.068	-0.101
41.35	0.985		0.053	0.014	0.011		-0.028		-0.056	-0.098
42.35	1.009		0.052	0.032	0.022		-0.023		-0.037	-0.147
44.85	1.069		0.033	0.016	0.022		-0.008		-0.042	-0.085
45.85	1.092		0.019	0.019	0.022					
46.85	1.116		0.049	0.018	0.027					

NORMAL ROWS

ROW ID		A	B	C	D	E	F	G	H
X		12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR		0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069
Y	Y/CR								
16.75	0.399			-0.342	-0.216	-0.152			
13.75	0.328		-0.252	-0.266	-0.221	-0.143			
10.75	0.256		-0.319	-0.299	-0.219	-0.143	-0.098	-0.147	
7.75	0.185	-0.362	-0.353	-0.312	-0.234	-0.151	-0.083		-0.044
6.75	0.161					-0.243	-0.145	-0.056	-0.037 -0.042
5.75	0.137	-0.396	-0.375	-0.319	-0.244	-0.143	-0.062	-0.051	-0.021
4.75	0.113					-0.241	-0.131	-0.059	-0.038 -0.018
4.25	0.101	-0.397	-0.384	-0.344					
3.75	0.089					-0.234	-0.142	-0.028	-0.023 -0.008
2.75	0.066					-0.235	-0.127	-0.020	-0.007 -0.009
1.75	0.042					-0.252	-0.147	-0.002	-0.001 -0.002
0.75	0.018					-0.249	-0.133	0.011	0.022 0.022
-0.25	-0.006							0.014	0.032 0.016
-1.25	-0.030					0.140	0.140	0.053	0.052 0.033
-2.25	-0.054					0.125	0.127	0.079	0.076 0.032
-2.75	-0.066			0.055	0.041				
-3.25	-0.077					0.109	0.122	0.074	0.064 0.039
-4.25	-0.101	0.094	0.055	0.038	0.093	0.104	0.065	0.069	0.041
-5.25	-0.125					0.084	0.090	0.067	0.059 0.034
-6.25	-0.149	0.105	0.057	0.044	0.109	0.084	0.063		-0.085
-9.25	-0.220		0.061	0.046	0.085	0.082	0.076		
-12.25	-0.292		0.066	0.051	0.066	0.071			
-15.25	-0.363			0.051	0.077	0.057			

RUN.SEQ
206.2

MACH	RN/L	RN	PT	P	TTR	Q	ALPHA	CONF
0.249	1.511	3.44	1947	1865	537.8	80.9	5.00	17

WING COEFFICIENTS

CNU	CNL	CN	CMU	CML	CM	CB	XCPU	XCPL	XCP	YCP	TAU	CF
0.293	0.132	0.425	0.0015	-0.0202	-0.0187	0.1805	24.48	40.37	29.40	42.49	0.000	0.00

WING SECTION COEFFICIENTS

2Y/B	CNUS	CNLS	CNS	CMUS	CMLS	CMS	XCPUS	XCPLS	XCPS	CNC/	CMC/
0.099	0.235	0.128	0.364	-0.0112	-0.0289	-0.0401	29.77	47.50	36.03	0.521	0.136
0.296	0.292	0.142	0.434	-0.0037	-0.0314	-0.0351	26.28	47.13	33.09	0.530	0.017
0.500	0.336	0.148	0.484	-0.0081	-0.0331	-0.0412	27.41	47.37	33.52	0.484	-0.082
0.697	0.357	0.136	0.493	-0.0134	-0.0303	-0.0437	28.75	47.27	33.87	0.388	-0.110
0.894	0.349	0.098	0.447	-0.0291	-0.0243	-0.0534	33.32	49.89	36.94	0.257	-0.085

WING LOWER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.504	0.504	0.504	0.504	0.504
0.01			0.489		
0.02			0.444		
0.03			0.375		
0.04			0.289		
0.05	0.245	0.282	0.256	0.240	0.224
0.10	0.196	0.170	0.197	0.151	0.144
0.15	0.137	0.114	0.165	0.154	0.121
0.20	0.090	0.133	0.160	0.127	0.069
0.30	0.051	0.098	0.099	0.118	-0.003
0.40	0.080	0.099	0.062	0.075	-0.016
0.50	0.073	0.067	0.028	0.046	0.033
0.55					
0.60	0.118	0.081	0.126	0.064	0.075
0.65			0.178		
0.70	0.151	0.132	0.176	0.186	0.171
0.75	0.150	0.194	0.171	0.210	0.170
0.80	0.138	0.210	0.171	0.230	0.130
0.85	0.180	0.222	0.216	0.204	0.142
0.90	0.155	0.171	0.183	0.147	0.172
0.95	0.136	0.108	0.167	0.103	0.131
1.00	0.061	0.095	0.075	0.023	-0.010

WING UPPER SURFACE COEFFICIENTS

2Y/B	0.099	0.296	0.500	0.697	0.894
X/C					
0	0.276	-0.005	-0.204	-0.170	0.053
0.003			-0.640		
0.006			-0.880	-0.850	
0.01	-0.505	-0.963	-1.077	1.059	-0.675
0.02	-0.733	-1.150	-1.205	-1.093	-0.730
0.03	-0.769	-1.073	-1.167	-1.051	-0.733
0.04	-0.736	-0.987	-1.101	-0.964	-0.699
0.05	-0.661	-0.919	-1.026	-0.999	-0.761
0.06	-0.598	-0.861	-0.900	-0.891	-0.641
0.08	-0.521	-0.758	-0.786	-0.808	-0.599
0.10	-0.505	-0.713	-0.743	-0.716	-0.554
0.125	-0.479	-0.598	-0.698	-0.659	-0.552
0.15	-0.459	-0.539	-0.641	-0.618	-0.596
0.175	-0.395	-0.471	-0.559	-0.592	-0.584
0.20	-0.360	-0.462	-0.512	-0.599	-0.535
0.225	-0.316	-0.440	-0.465	-0.573	-0.518
0.25	-0.307	-0.434	-0.458	-0.501	-0.499
0.30	-0.314	-0.353	-0.447	-0.476	-0.527
0.35	-0.298	-0.303	-0.413	-0.422	-0.529
0.40	-0.246	-0.265	-0.336	-0.396	-0.499
0.45	-0.218	-0.263	-0.300	-0.412	-0.407
0.50	-0.187	-0.256	-0.251	-0.370	-0.358
0.55	-0.193	-0.247	-0.246	-0.281	-0.302
0.60	-0.192	-0.180	-0.244	-0.247	-0.275
0.65	-0.183	-0.153	-0.220	-0.191	-0.279
0.70	-0.129	-0.107	-0.150	-0.180	-0.228
0.75	-0.098	-0.103	-0.111	-0.170	-0.147
0.80	-0.059	-0.101	-0.071	-0.144	-0.114
0.85	-0.053	-0.090	-0.050	-0.041	-0.057
0.90	-0.034	-0.006	-0.047	-0.007	-0.040
0.95	-0.011	0.046	-0.005	0.055	-0.044
1.00	0.061	0.095	0.075	0.023	-0.010

WALL TURNTABLE STATIC PRESSURE COEFFICIENTS

CHORDWISE ROWS

ROW ID	1A	1B	2	3	4A	4B	5A	5B	6
Y	-2.75	-1.25	-0.25	0.75	4.25	3.75	7.75	6.75	10.75
Y/CR	-.066	-.030	-.006	.018	.101	.089	.185	.161	.256

X	X/CR									
10.35	0.247	0.093				-0.328				
11.35	0.270	0.044				-0.234				
12.35	0.294					-0.268		-0.167		
14.35	0.342	0.078				-0.249				
15.35	0.366	0.085				-0.197				-0.120
16.35	0.390	0.126				-0.143				-5.545
17.35	0.413	0.177				-0.279		-0.247		-0.163
18.35	0.437	0.023				-0.191				-0.117
19.35	0.461	0.111				-0.231				-0.061
20.35	0.485	0.065				-0.214				-0.210
22.35	0.533	0.076				-0.152				-0.107
23.35	0.556	0.126				-0.094				-0.152
24.35	0.580	0.192				-0.230		-0.156		-0.141
25.35	0.604	0.029				-0.128				-0.096
26.35	0.628	0.131				-0.167				-0.035
27.35	0.652	0.098				-0.156				
30.35	0.723									-0.066
31.35	0.747		0.143		-0.115		-0.066	-0.059	-0.009	-0.100
32.35	0.771		0.193		-0.053		0.002		-0.146	-0.087
33.35	0.795		0.257		0.004		-0.150		-0.042	-0.035
34.35	0.818		0.094		-0.135		-0.042		-0.087	0.030
35.35	0.842		0.195		-0.033		-0.074		-0.077	-0.140
36.35	0.866		0.130		-0.057		-0.053		-0.016	-0.020
37.35	0.890		0.121		-0.027		0.019		0.049	-0.057
38.35	0.914		0.155		0.034		0.069		-0.097	-0.038
39.35	0.938		0.207	0.007	0.106		-0.065		0.003	0.008
40.35	0.961		0.036	0.029	-0.037		0.036		-0.021	0.062
41.35	0.985		0.116	0.101	0.079		0.001		-0.006	-0.093
42.35	1.009		0.083	0.166	0.042		0.018		0.049	-0.027
44.85	1.069		0.070	0.004	0.057		0.073		0.112	-0.023
45.85	1.092		0.120	0.090	0.090					
46.85	1.116		0.162	0.042	0.158					

NORMAL ROWS

ROW ID	A	B	C	D	E	F	G	H
X	12.35	17.35	24.35	31.35	36.35	41.35	42.35	44.85
X/CR	0.294	0.413	0.580	0.747	0.866	0.985	1.009	1.069

Y	Y/CR								
16.75	0.399			-0.052	-0.176	-0.018			
13.75	0.328		-0.113	-0.074	0.013	-0.120			
10.75	0.256		-0.163	-0.141	-0.100	-0.020	-0.093	-0.027	
7.75	0.185	-0.167	-0.247	-0.156	-0.059	0.036	-0.076		-0.010
6.75	0.161				-0.009	-0.016	-0.006	0.049	0.112
5.75	0.137	-0.244	-0.120	-0.182	-0.100	-0.014	0.093	0.045	-0.001
4.75	0.113				-0.115	-0.055	0.047	0.016	0.066
4.25	0.101	-0.268	-0.279	-0.230					
3.75	0.089				-0.066	-0.053	0.001	0.018	0.073
2.75	0.066				-0.083	-0.074	0.016	0.014	-0.041
1.75	0.042				-0.178	-0.012	0.022	0.075	0.151
0.75	0.018				-0.115	-0.057	0.079	0.042	0.057
-0.25	-0.006						0.101	0.166	0.004
-1.25	-0.030				0.143	0.130	0.116	0.083	0.070
-2.25	-0.054				0.172	0.134	0.195	0.195	0.080
-2.75	-0.066		0.177	0.192					
-3.25	-0.077				0.115	0.170	0.036	0.036	0.120
-4.25	-0.101	0.042	0.070	0.023	0.128	0.216	0.131	0.131	0.177
-5.25	-0.125				0.163	0.055	0.085	0.085	0.025
-6.25	-0.149	0.131	0.085	0.131	0.214	0.154	0.098		0.010
-9.25	-0.220		0.131	0.076	0.040	0.085	0.148		
-12.25	-0.292		0.184	0.087	0.129	0.096			
-15.25	-0.363			0.131	0.083	0.131			

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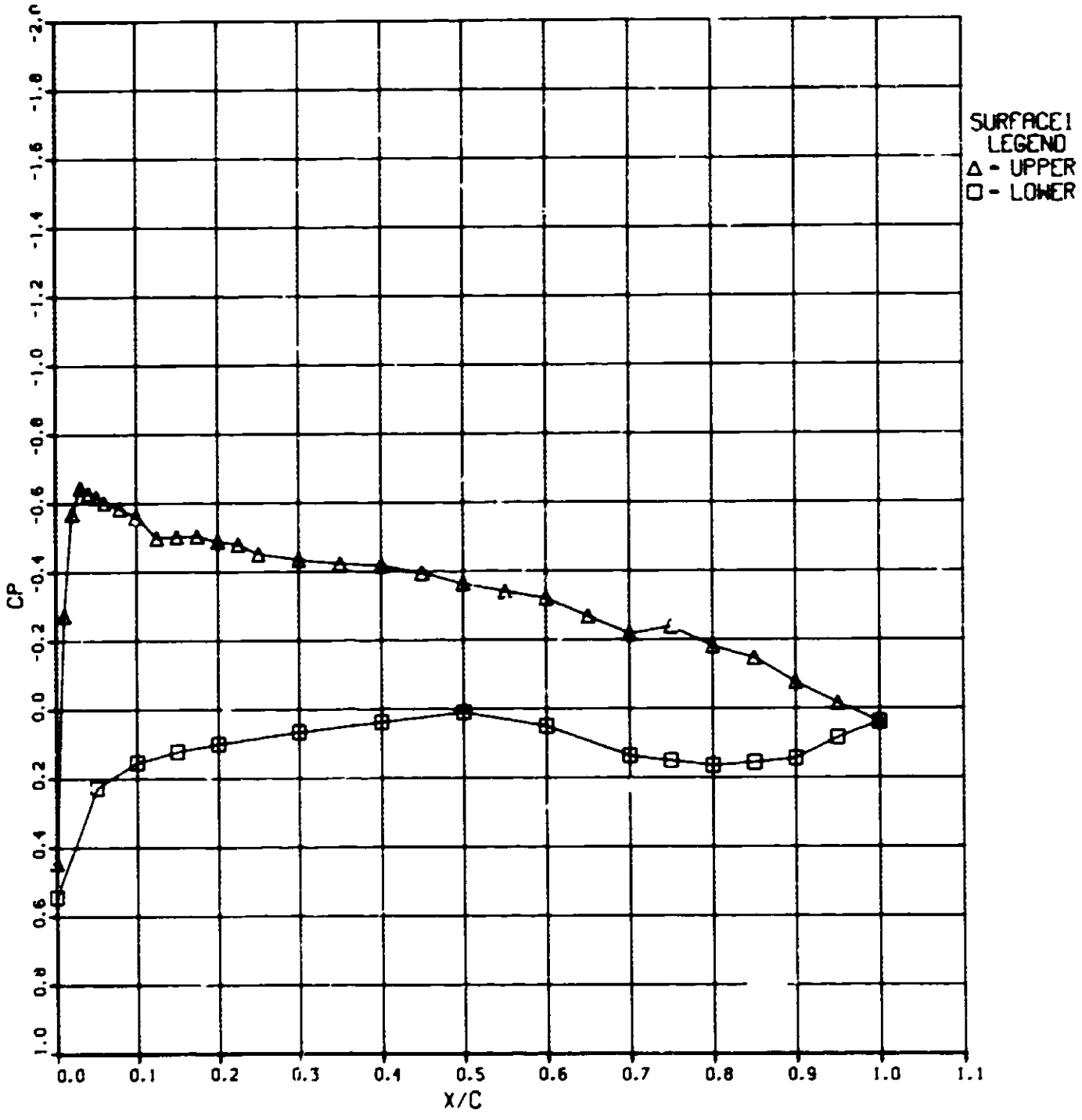
NASA AMES RESEARCH CENTER

FRICK

STOP 10

(000) 000-0000

356-1-66 165.00: 1.00 CONF-17 MACH-0.821 RN-2.990 PT-1500 ALPHA- 5.00



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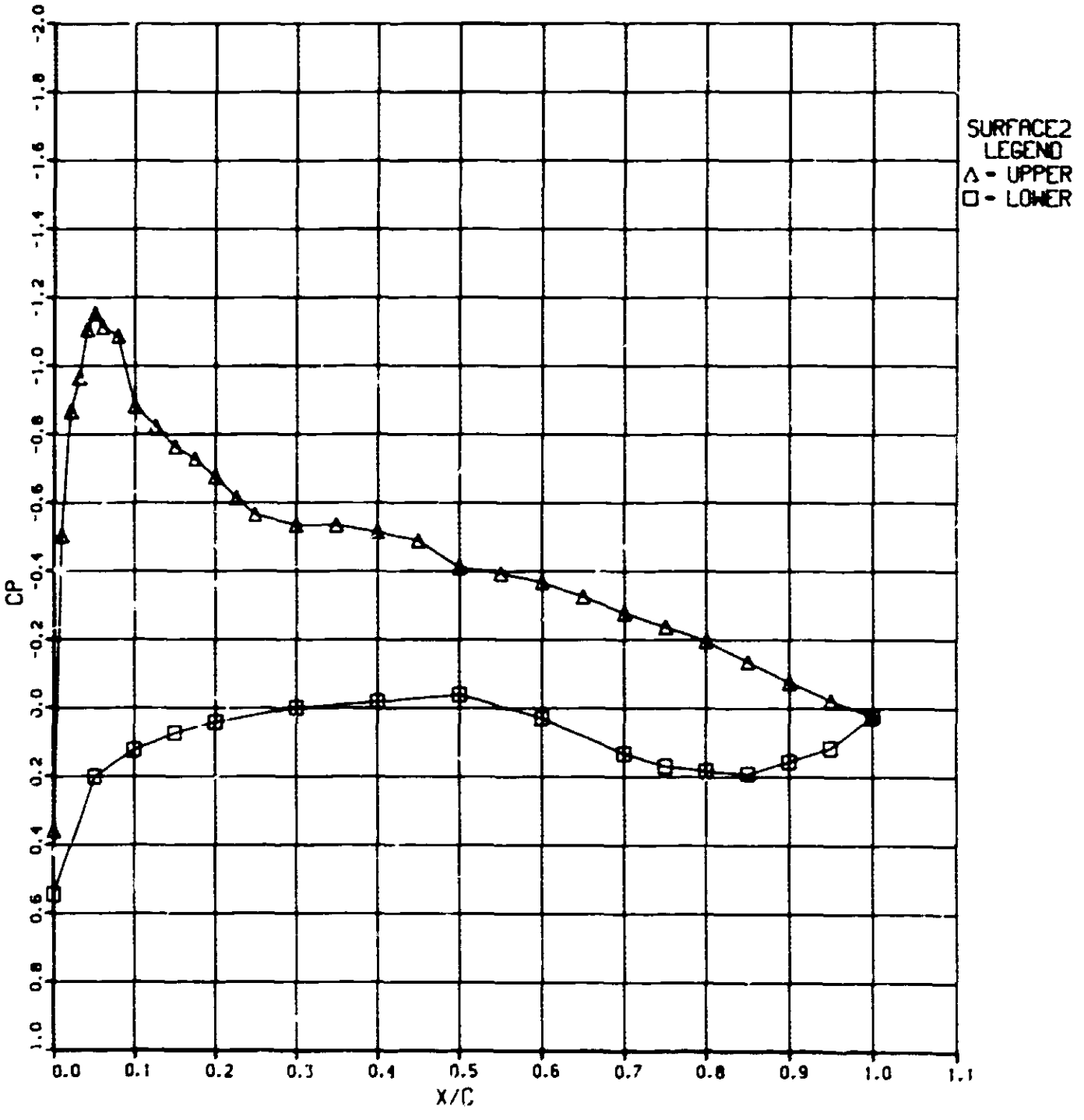

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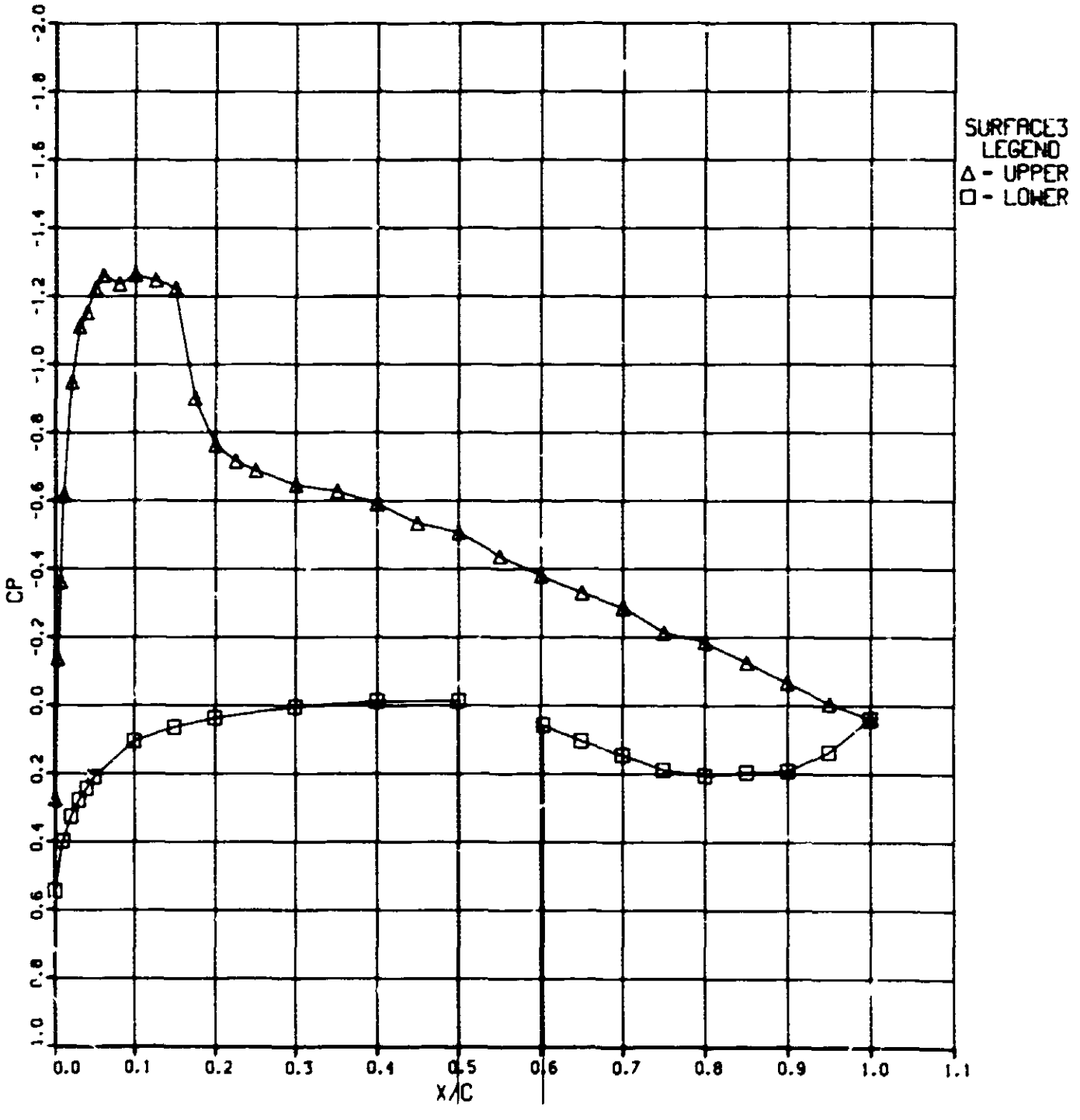

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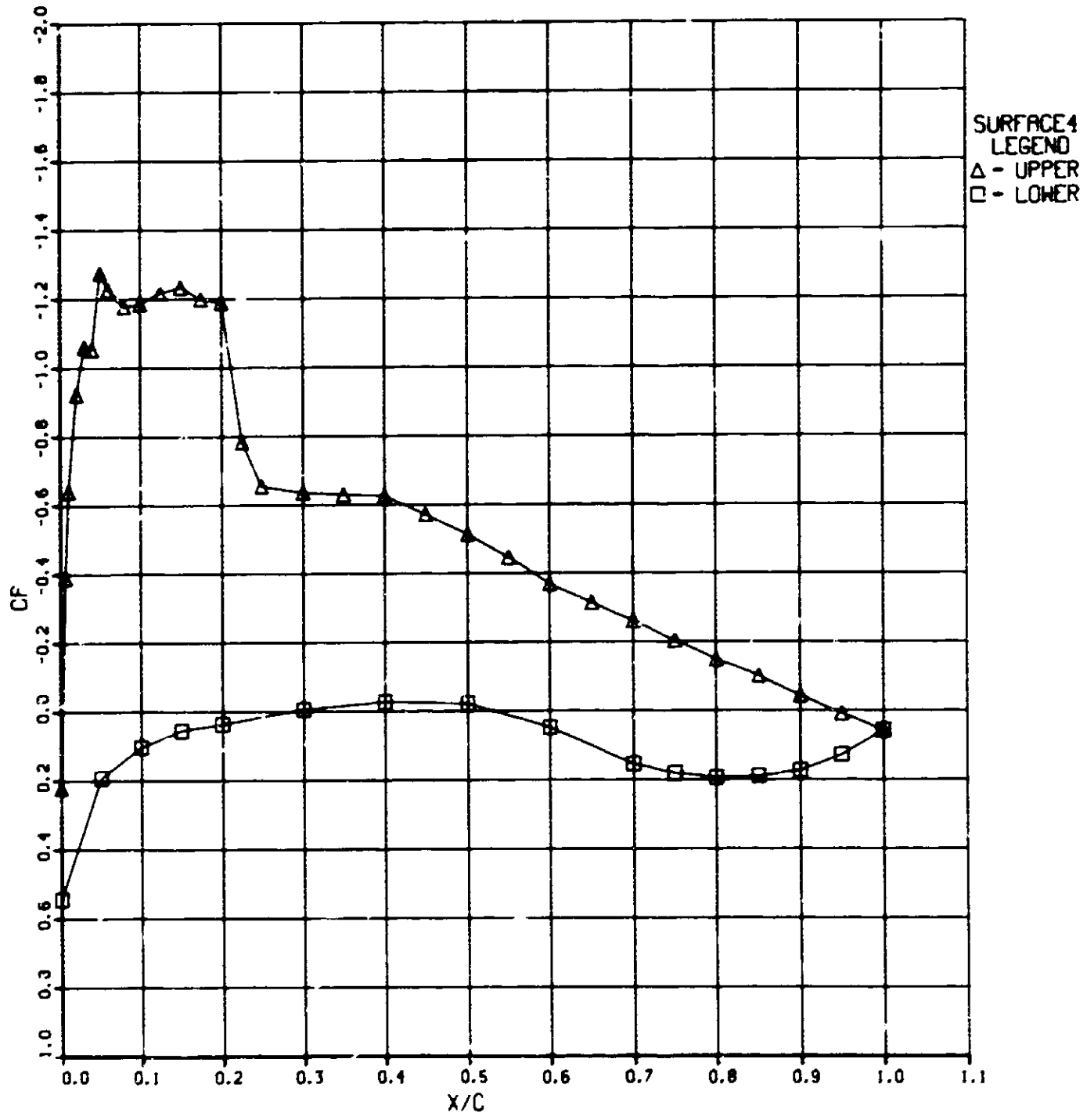

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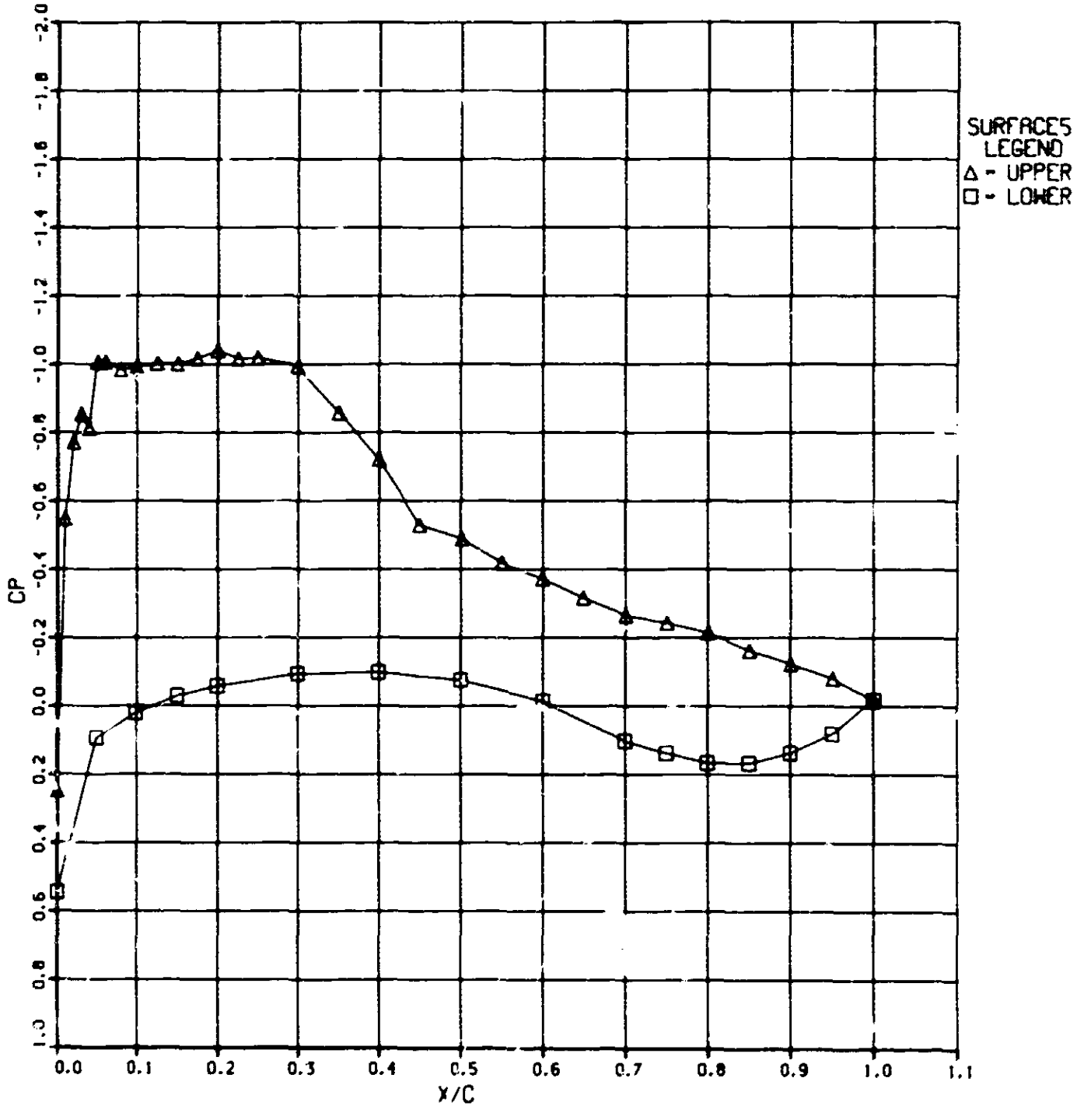

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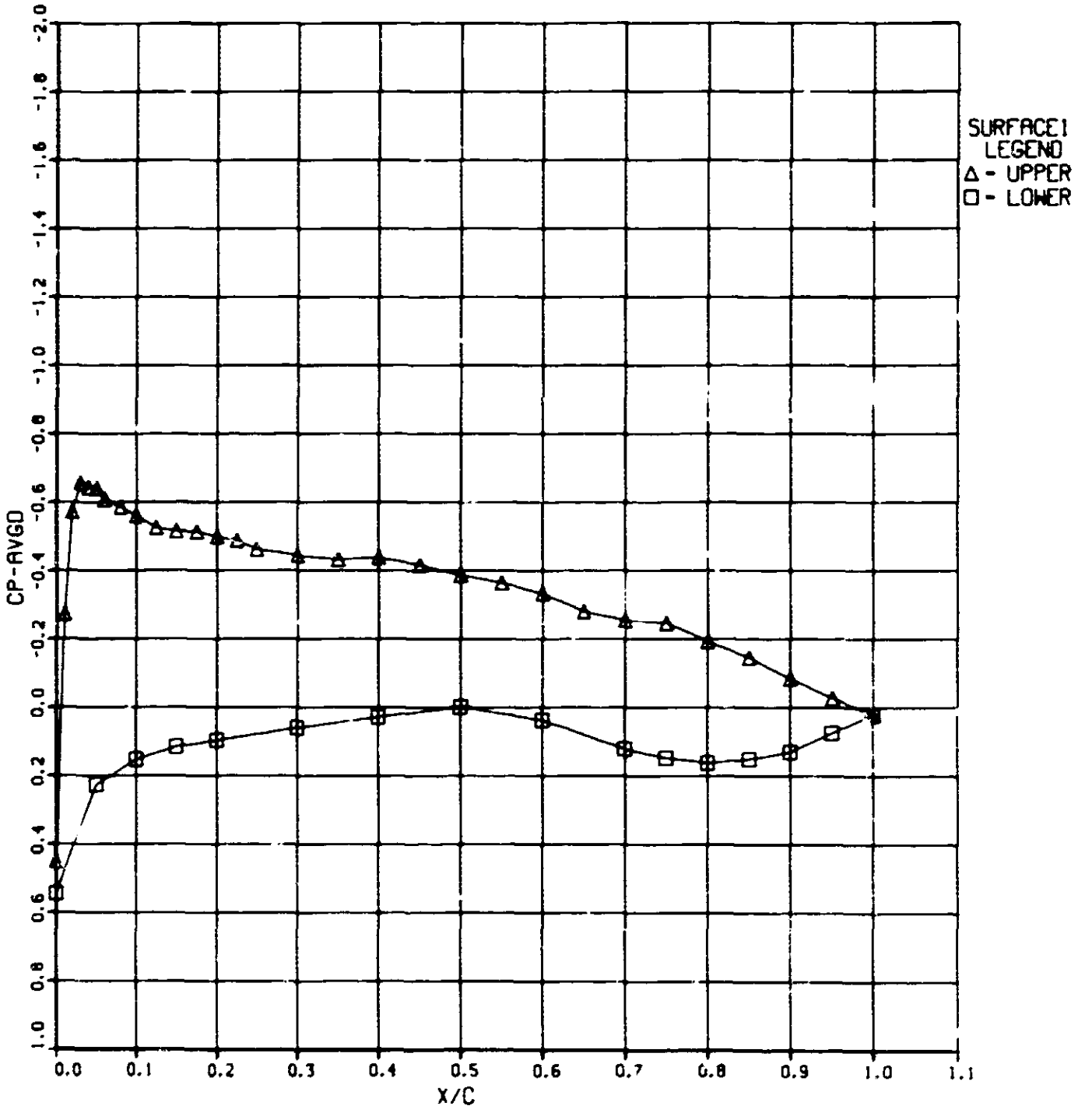

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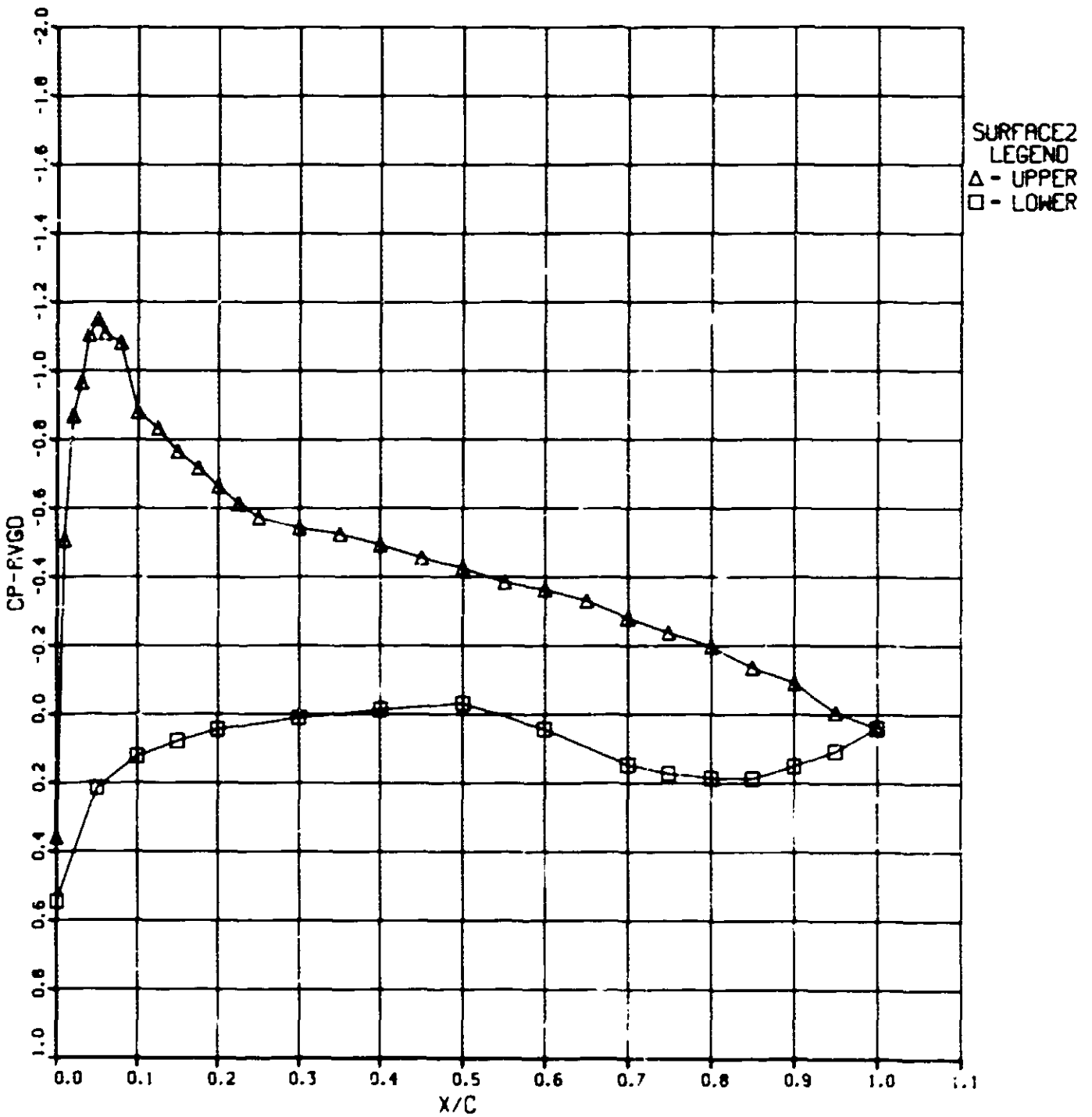

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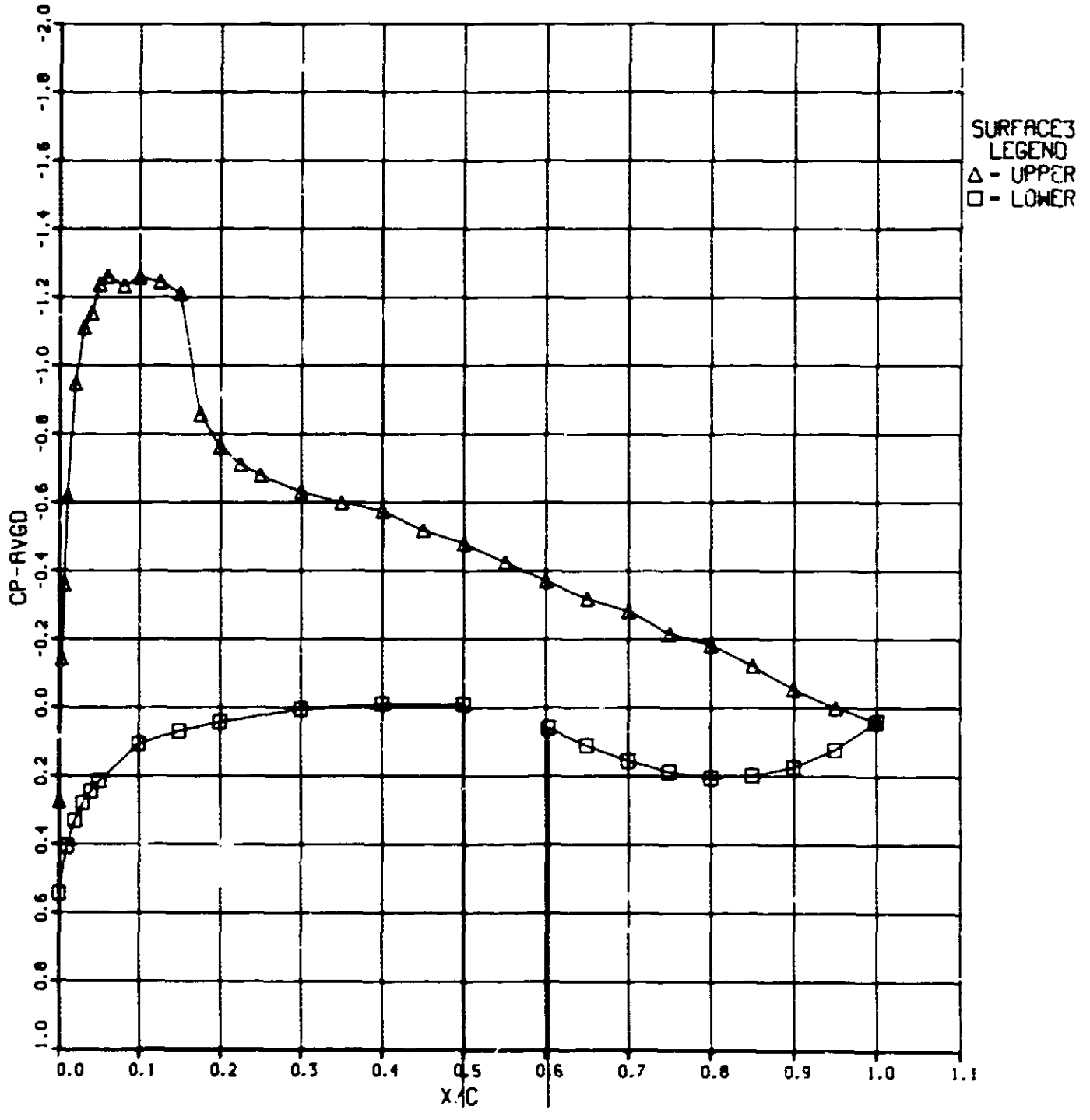

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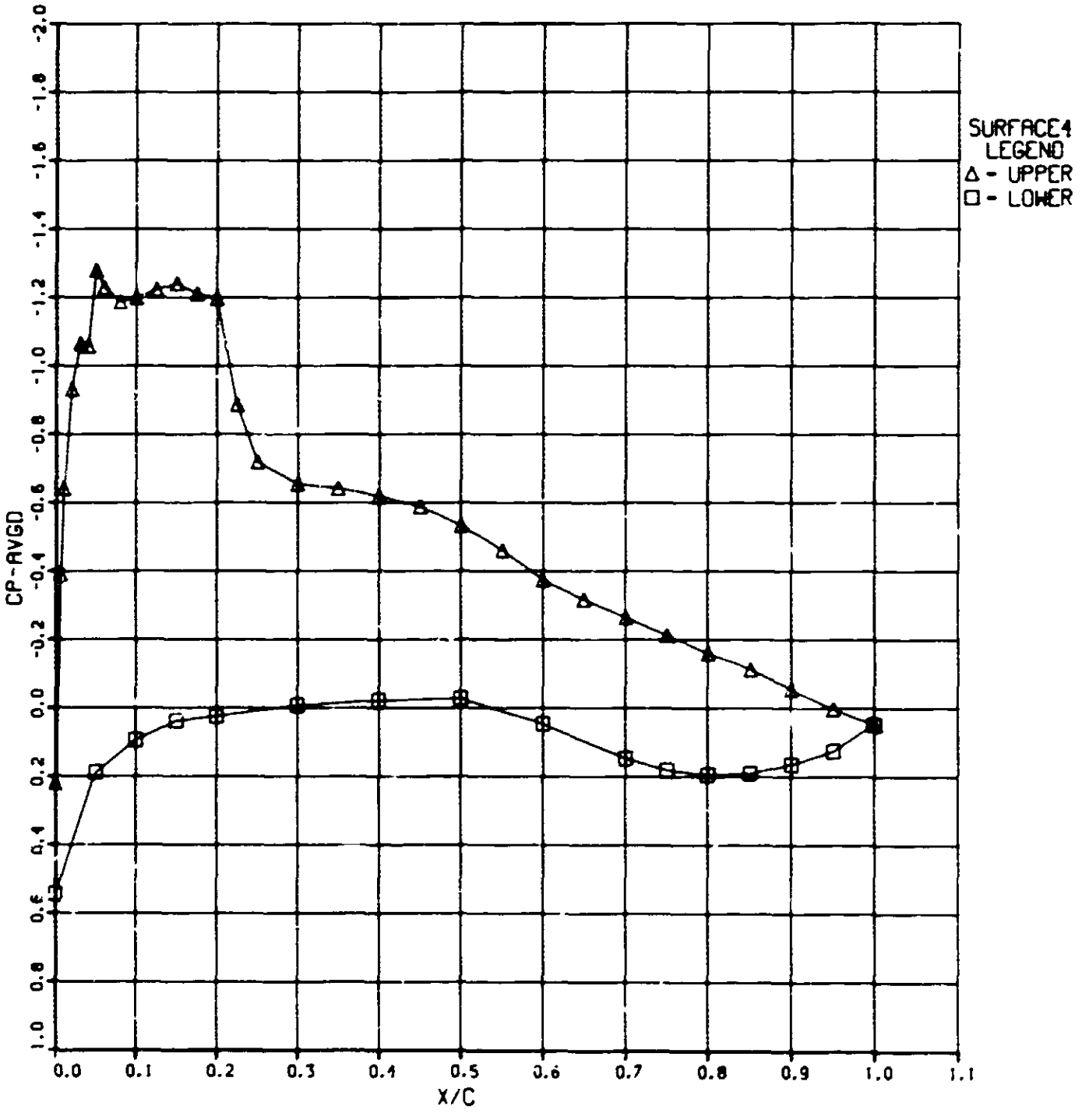

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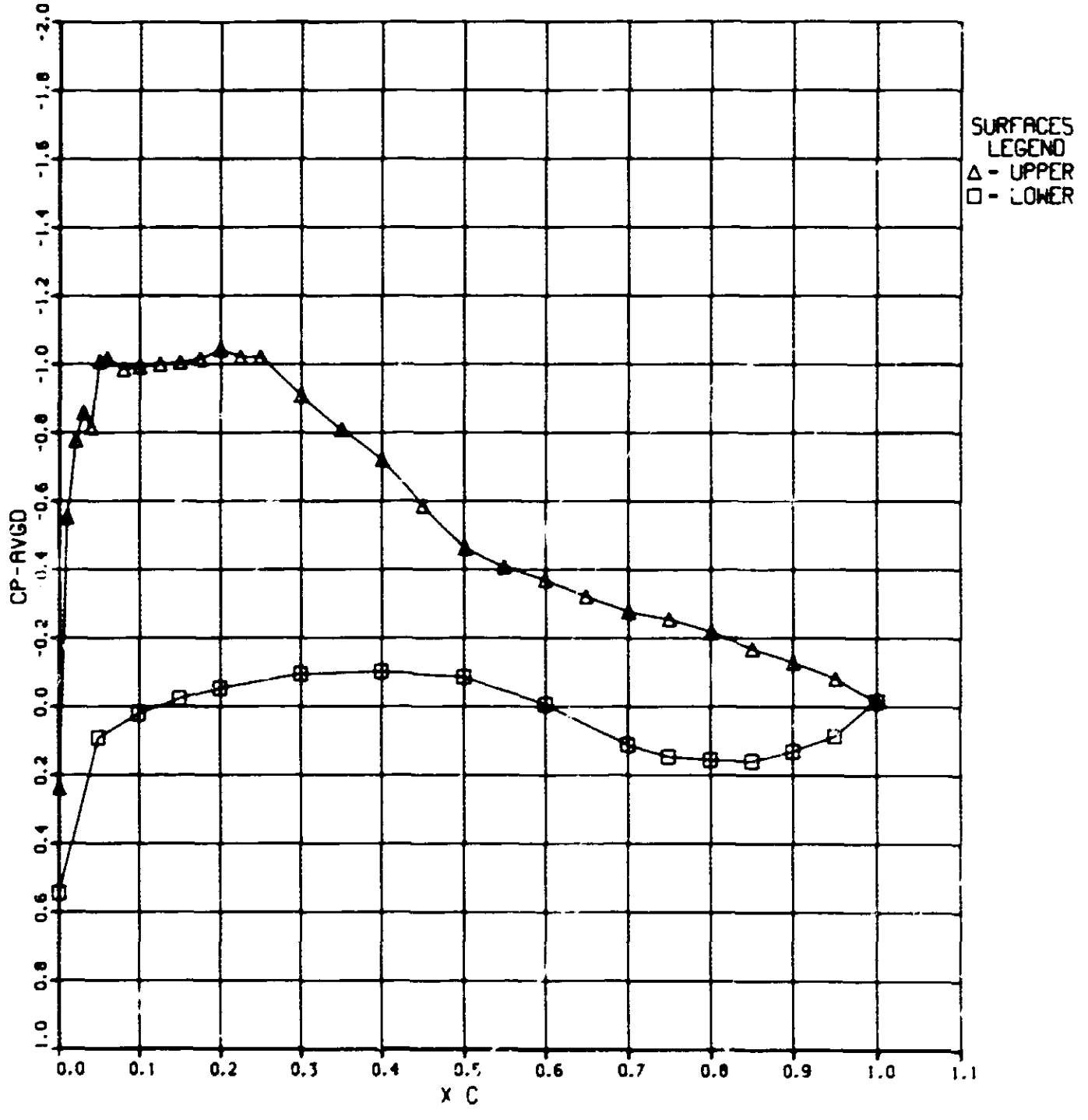

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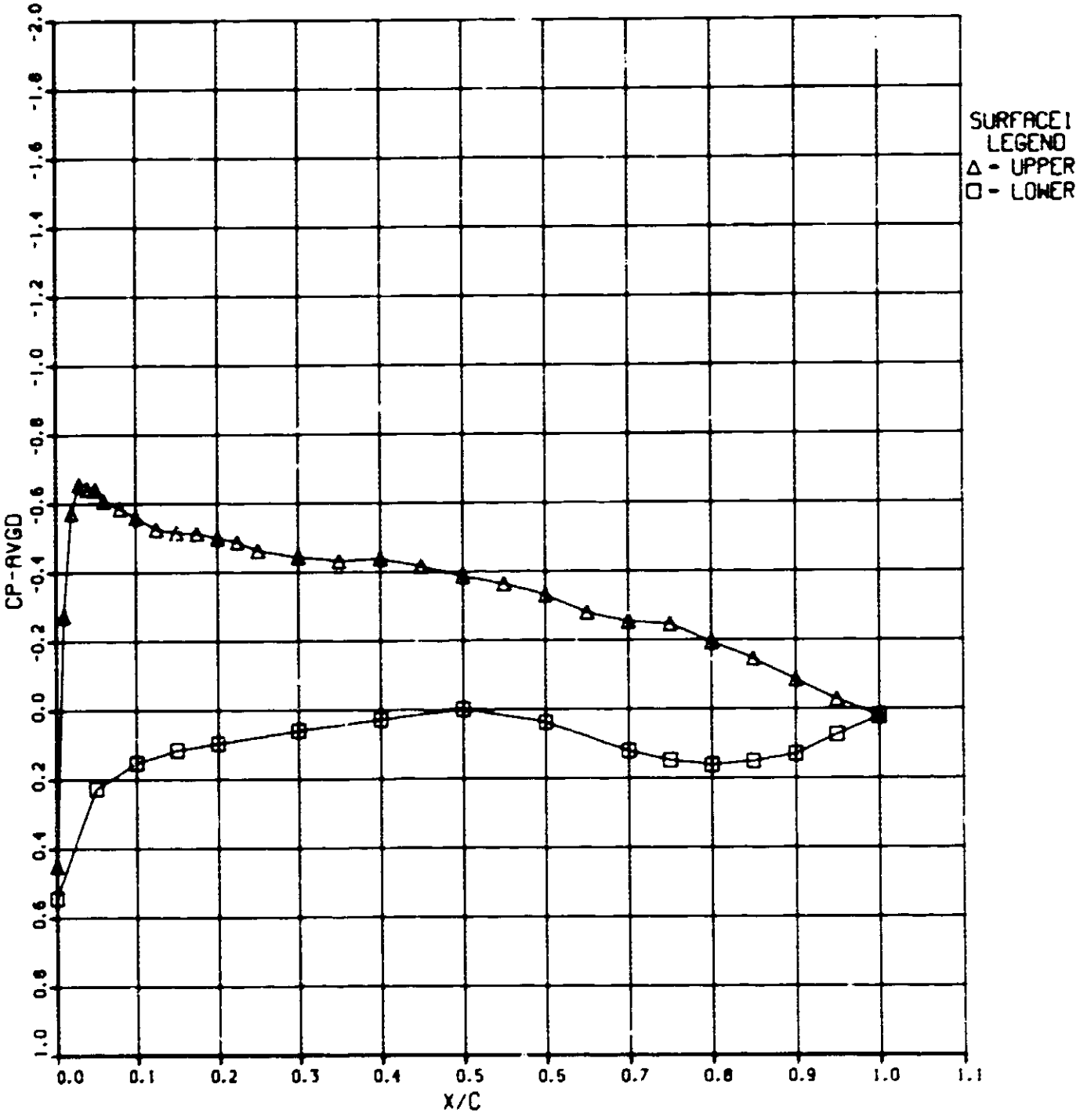

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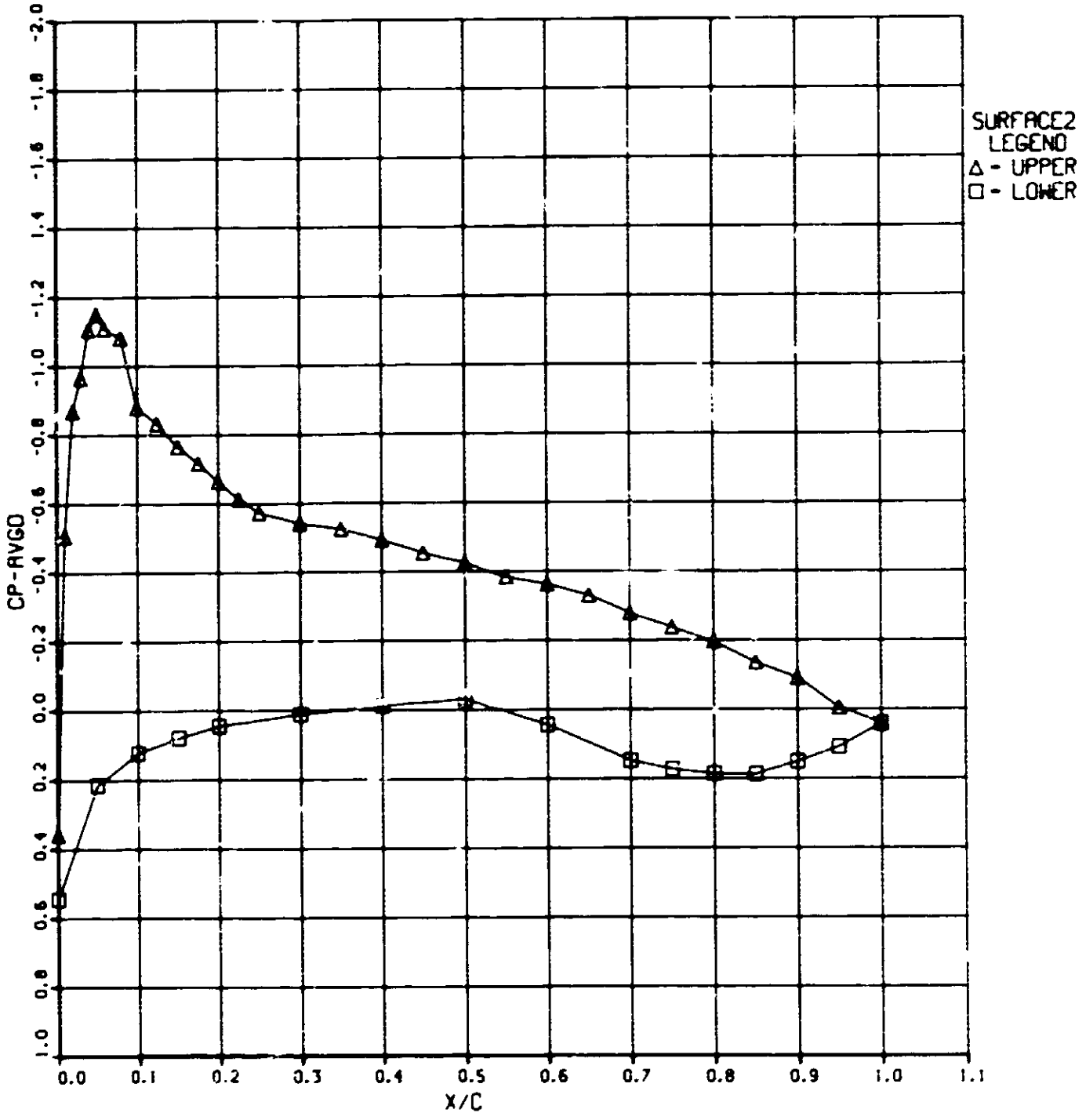

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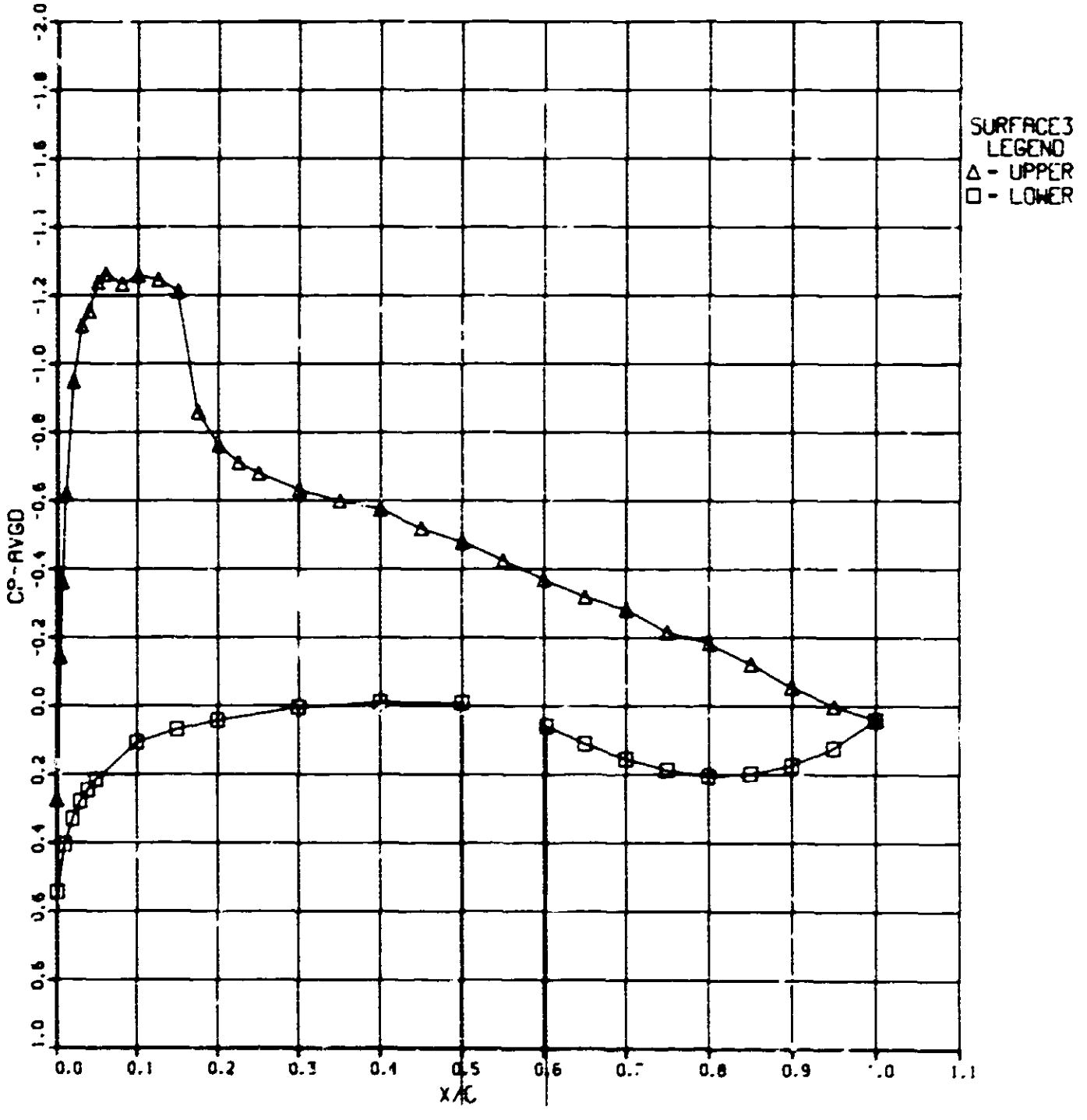

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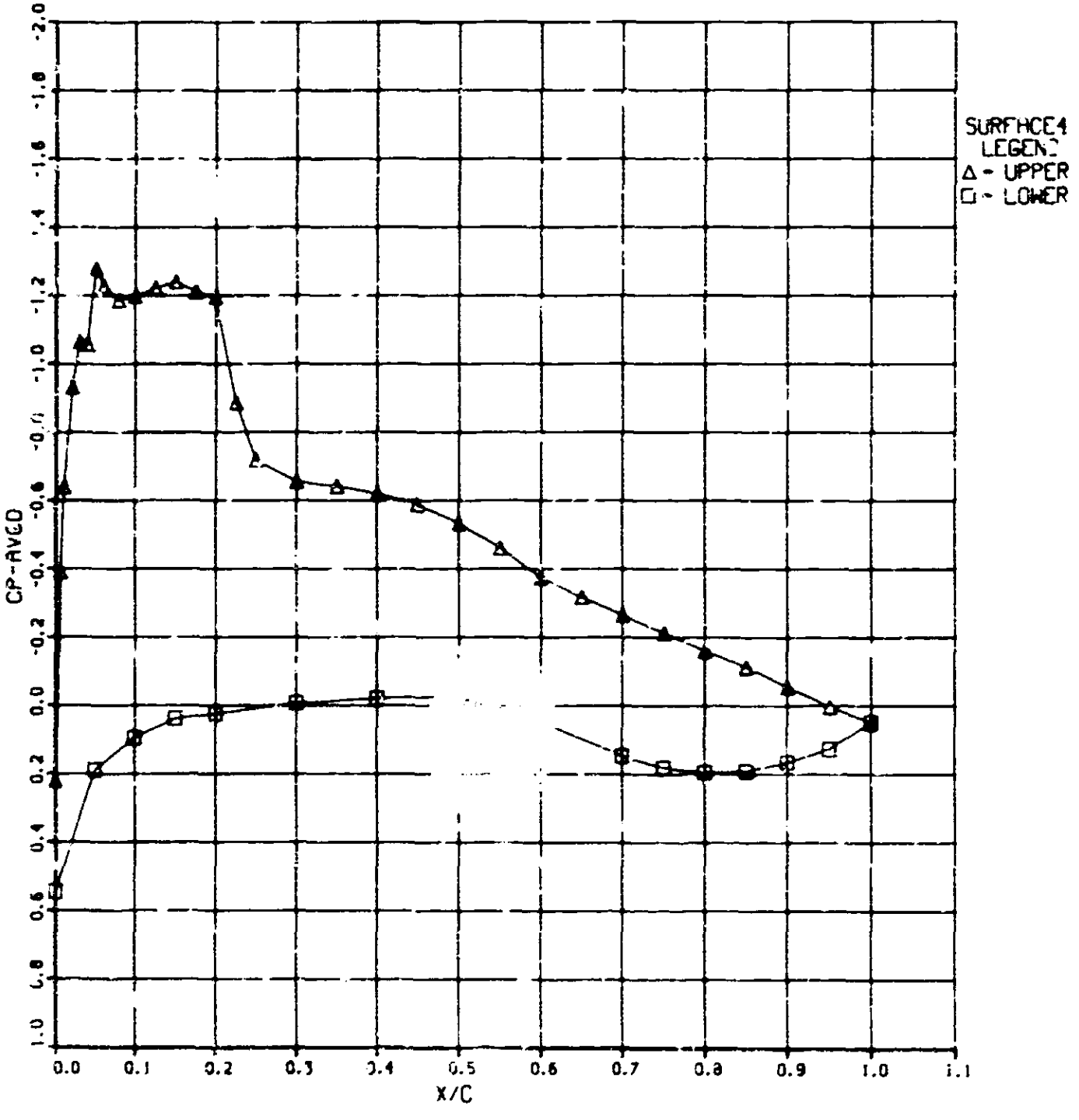

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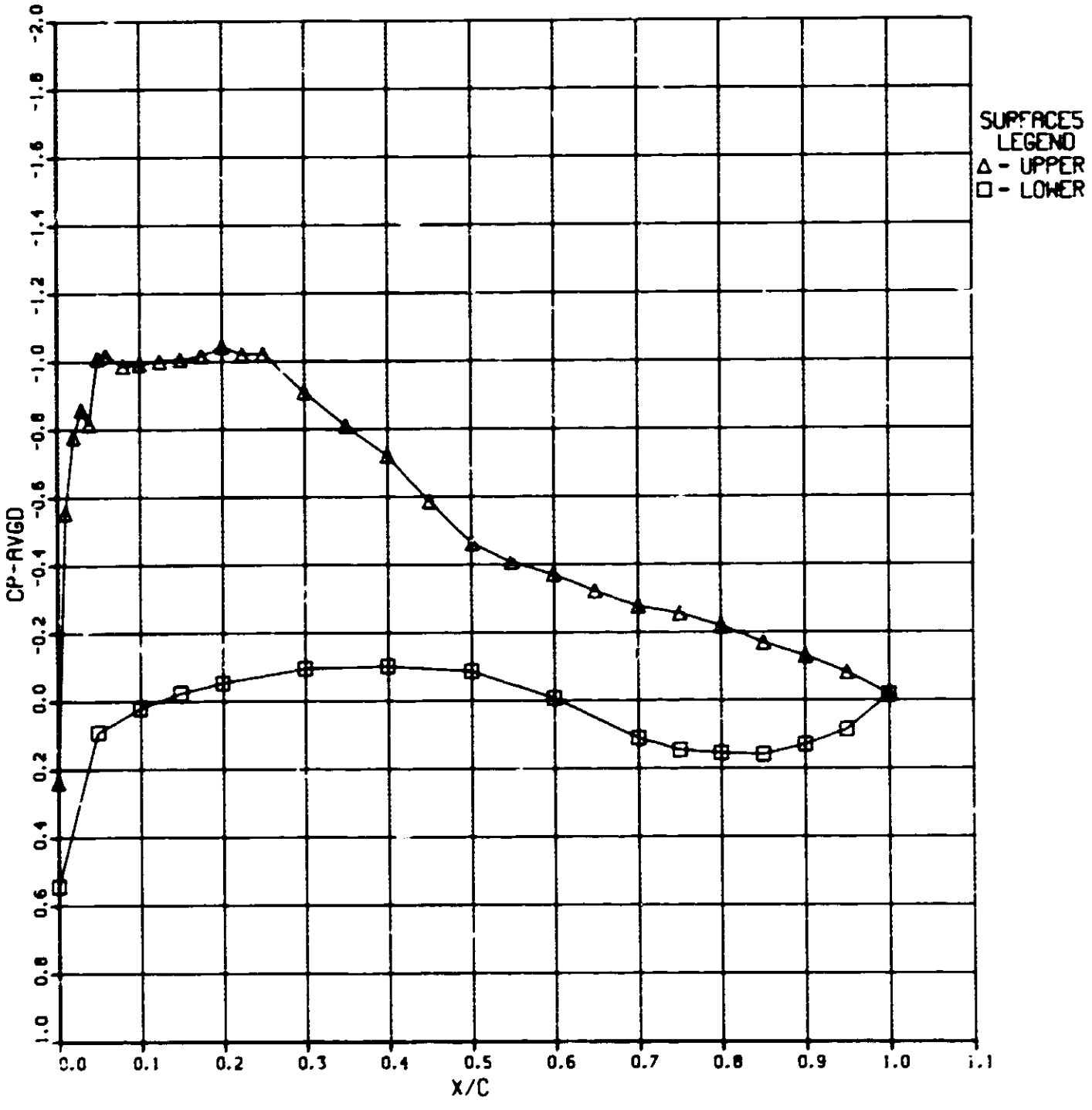

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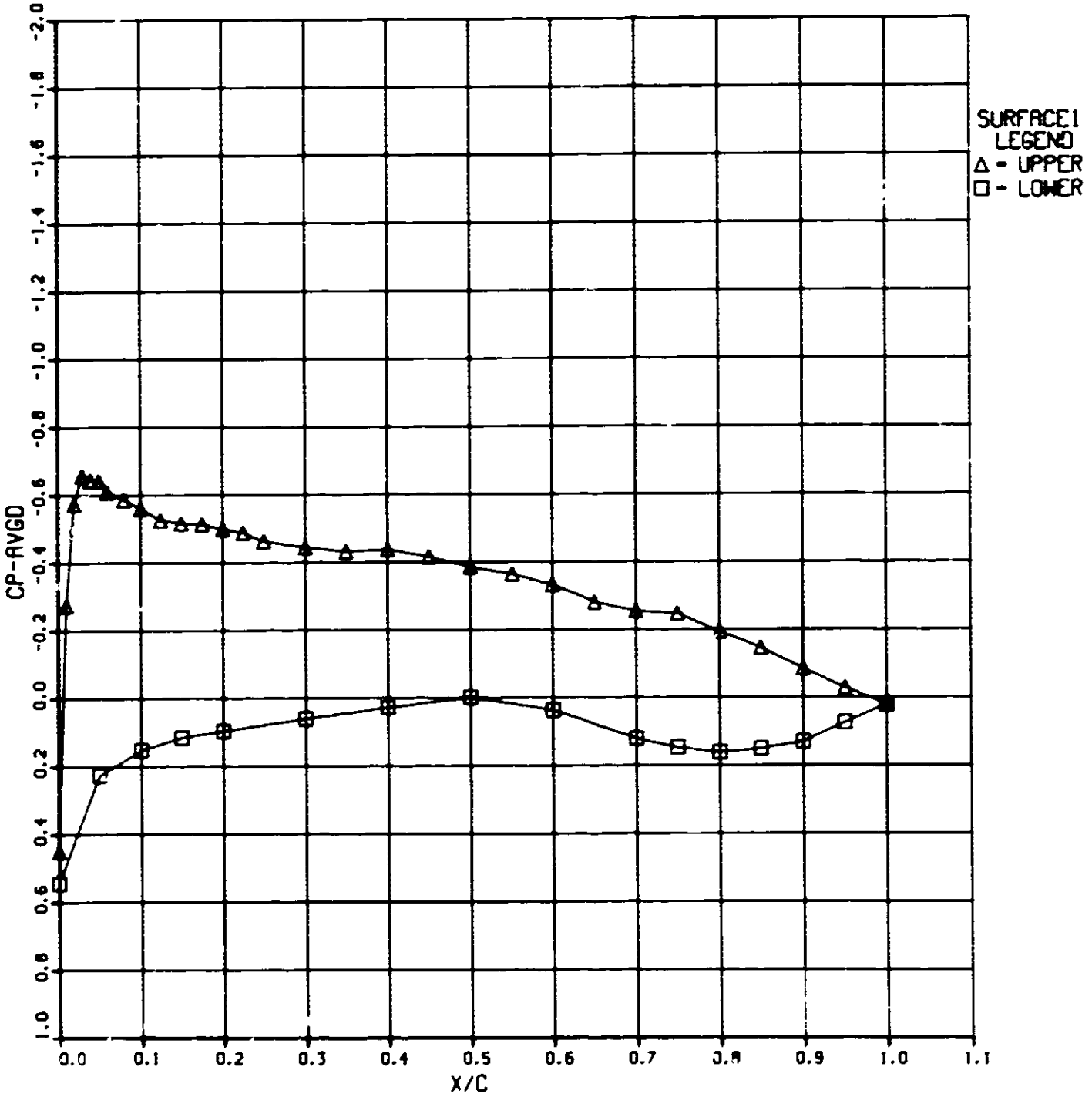

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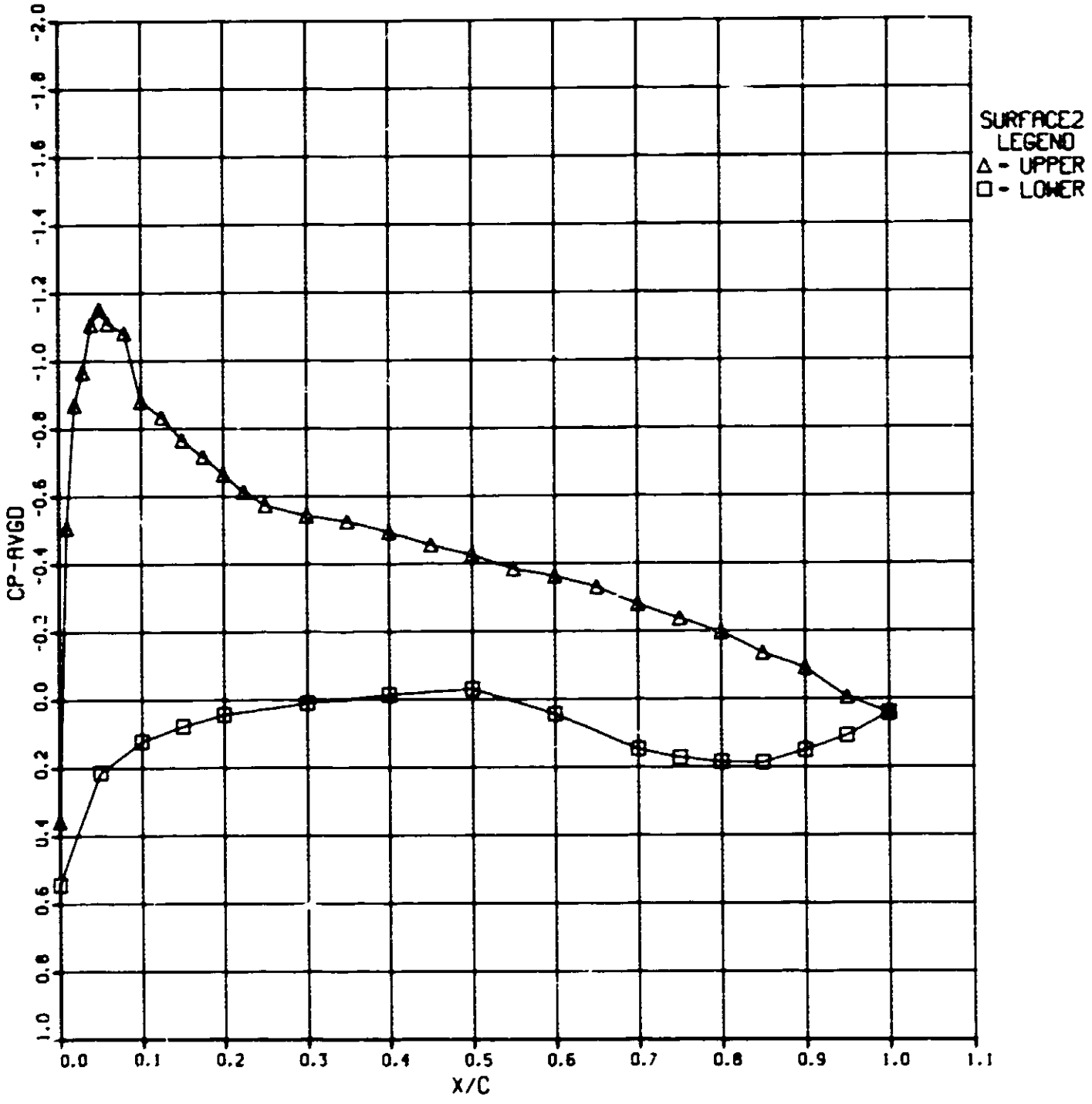

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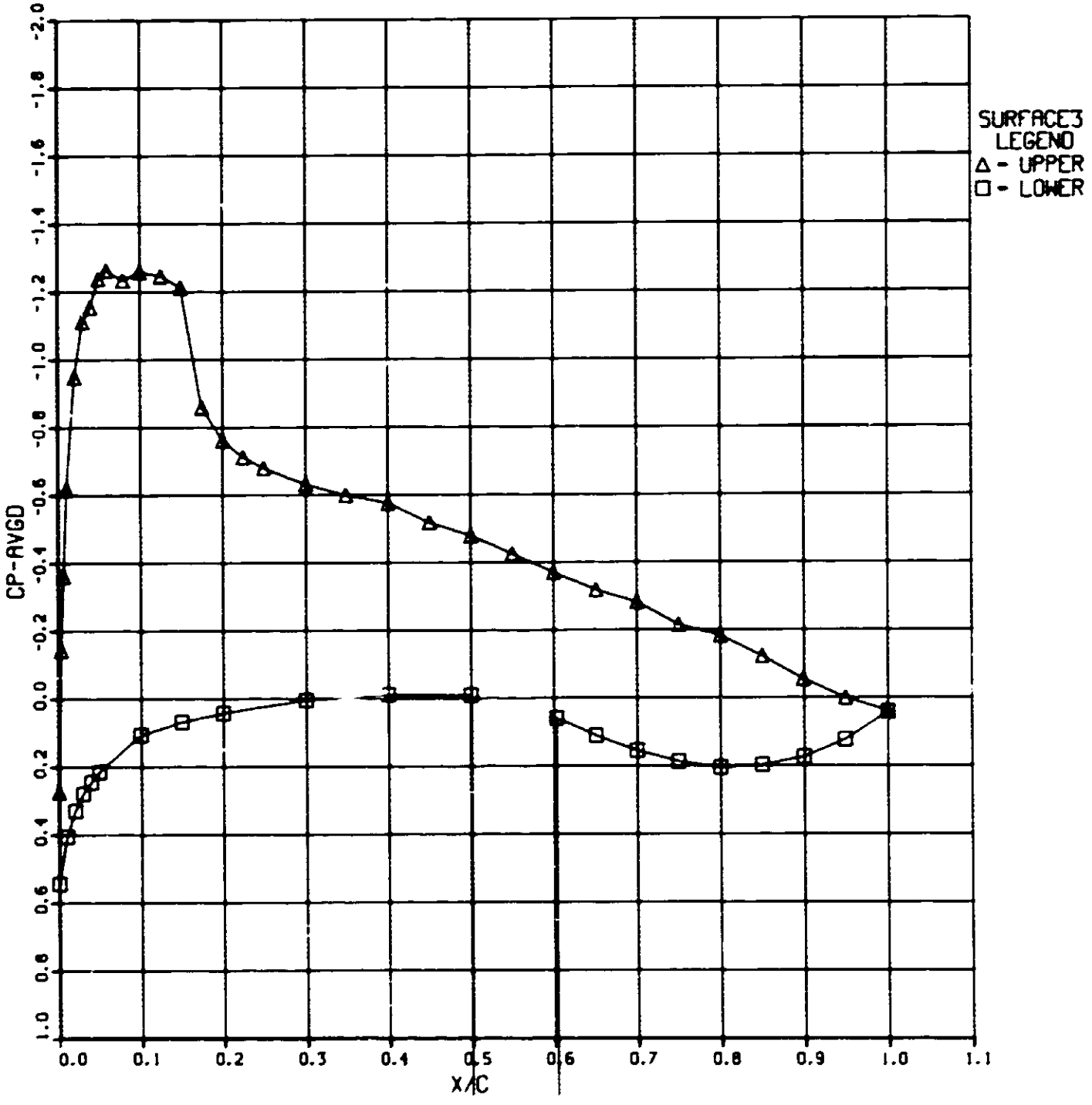

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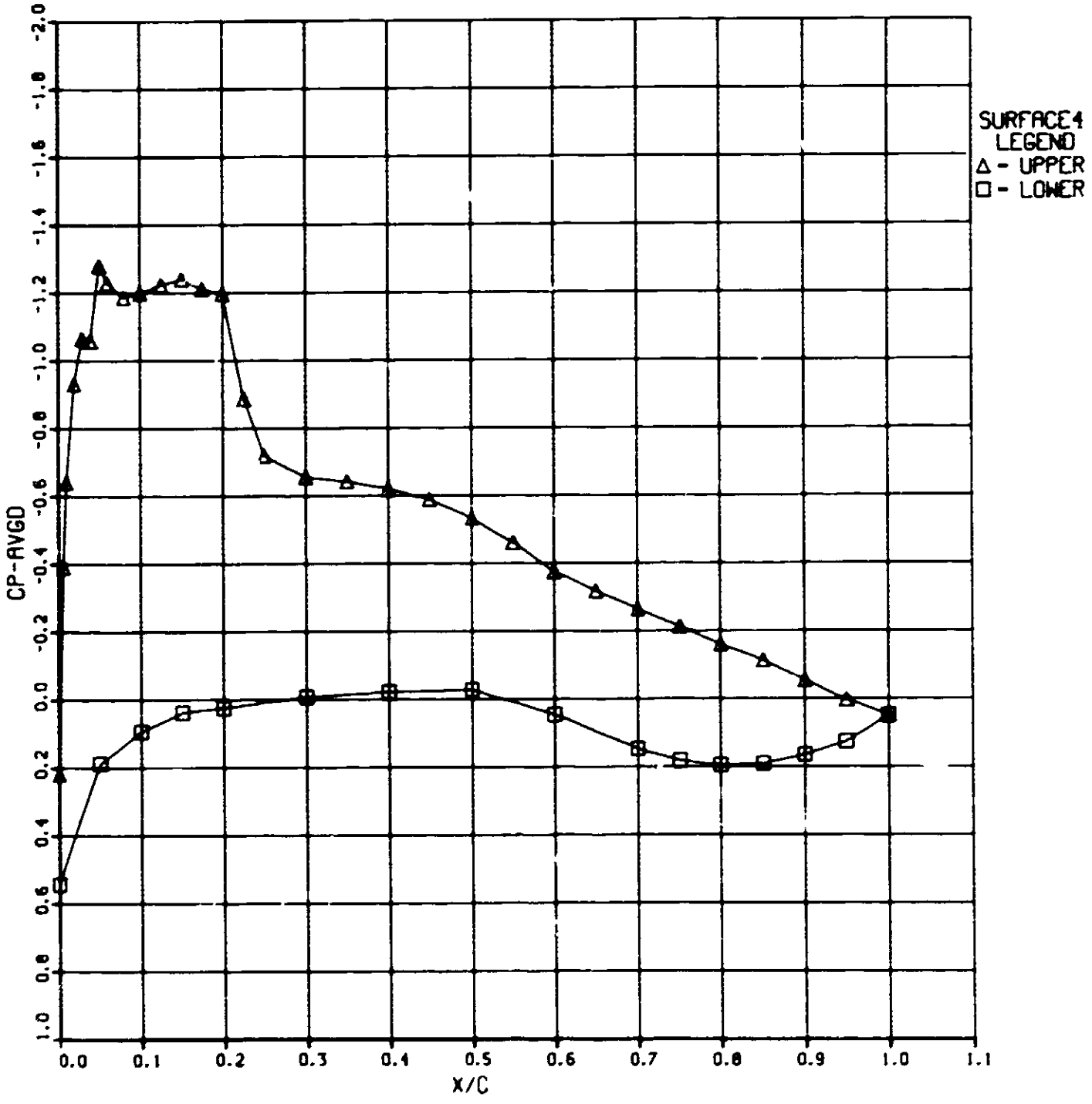

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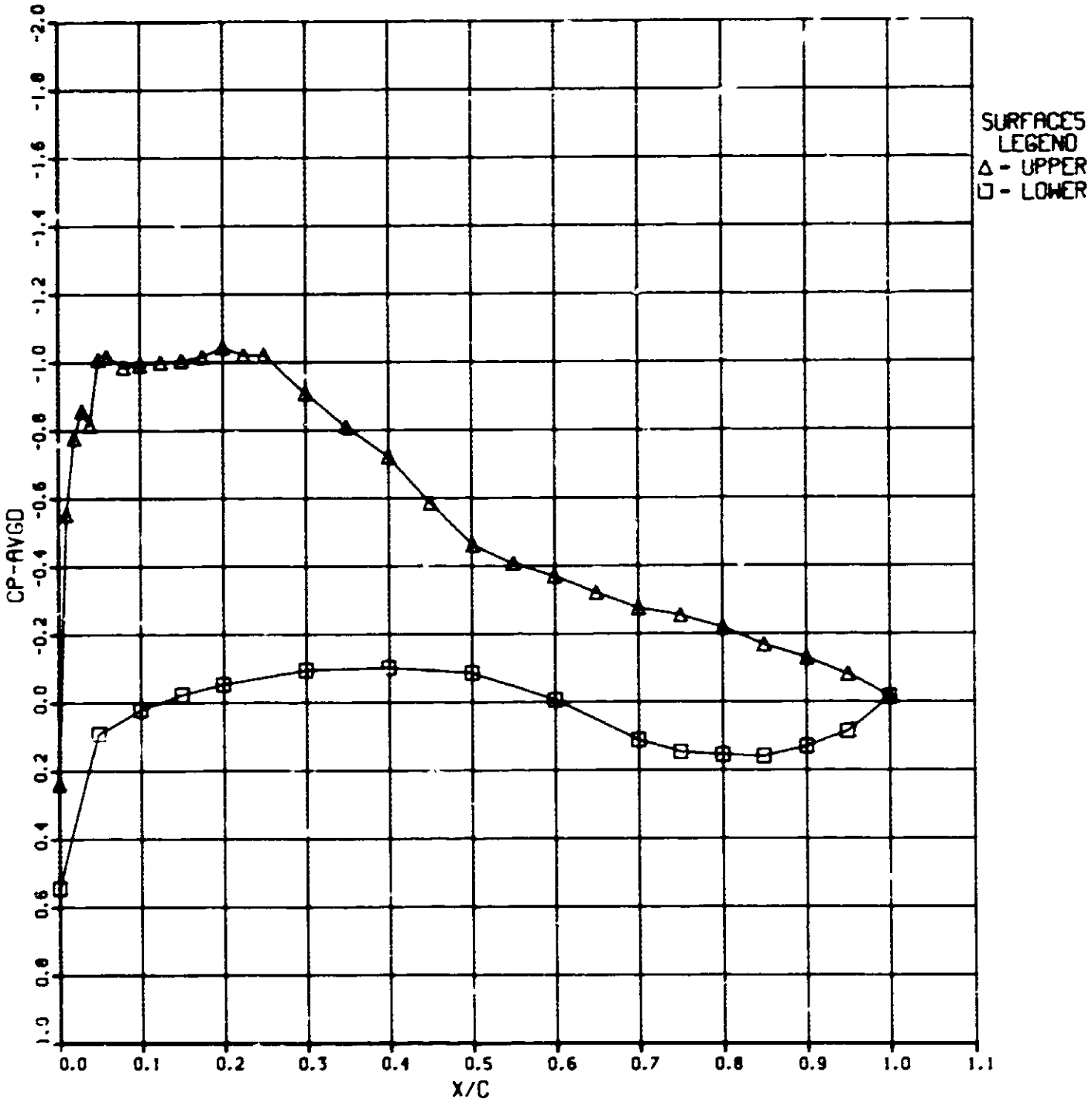
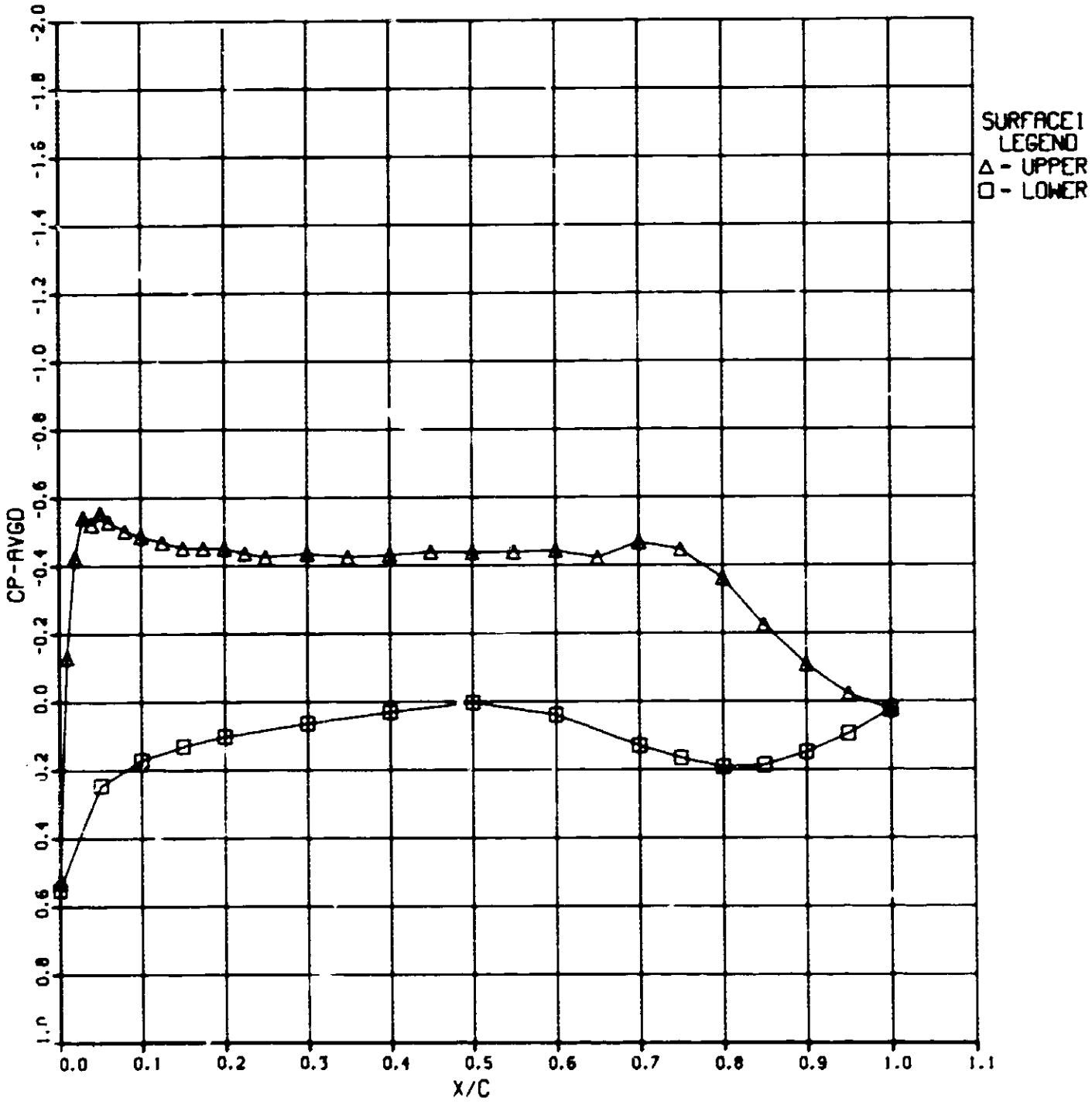


356-1-66 166.00: 3.00 CONF-17 MACH-0.893 RN-1.347 PT-2163 ALPHA- 5.00



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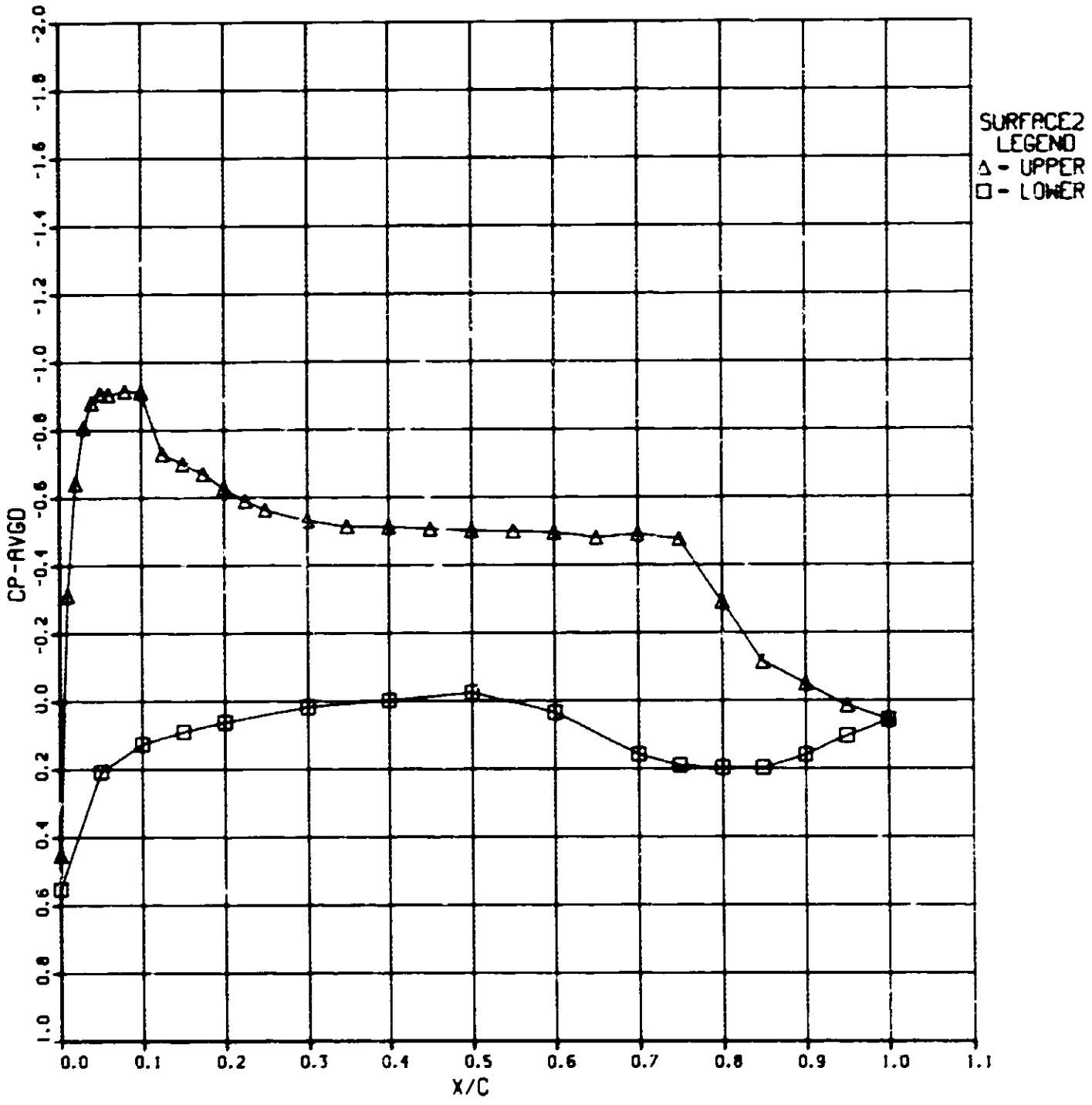

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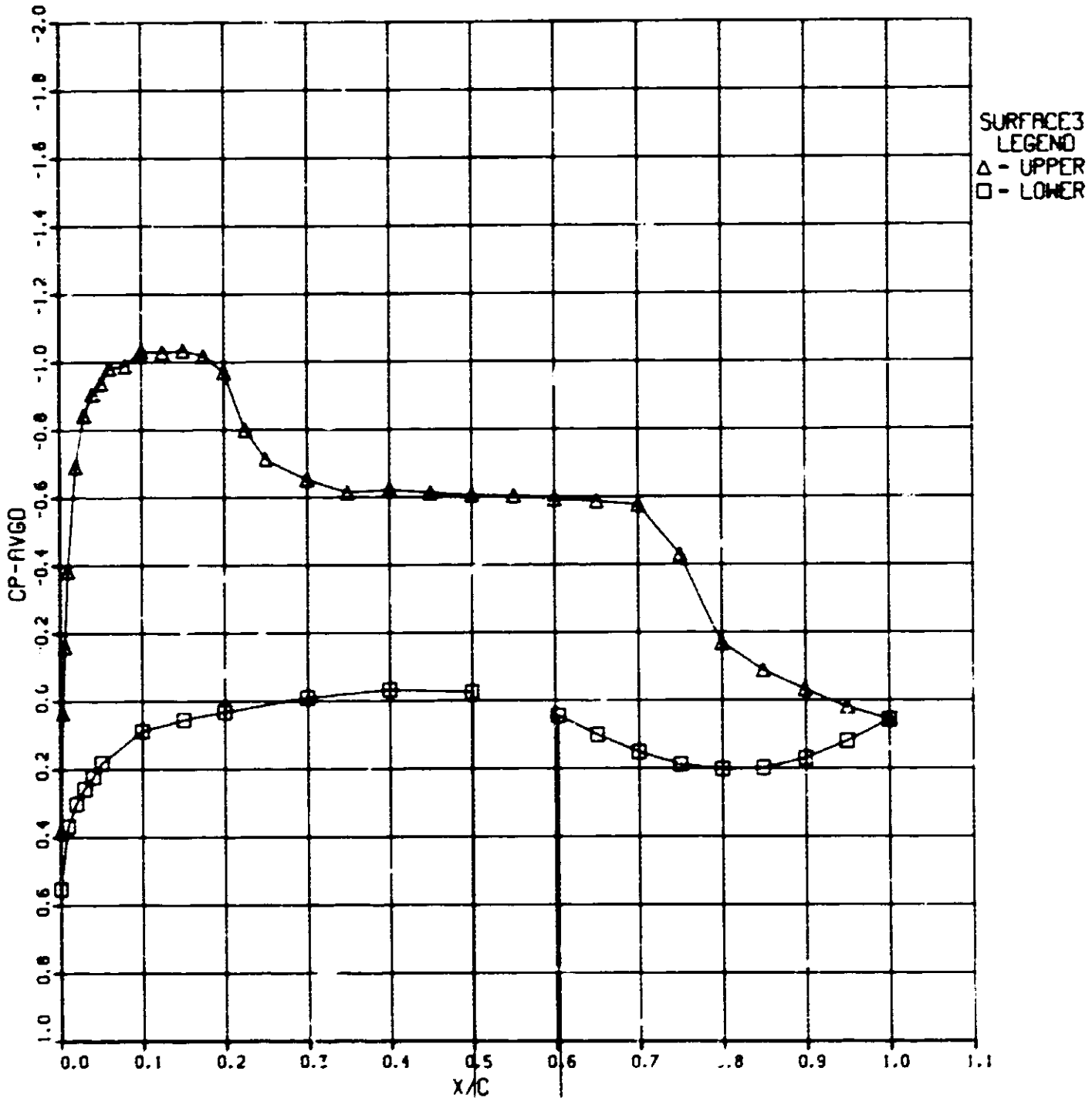
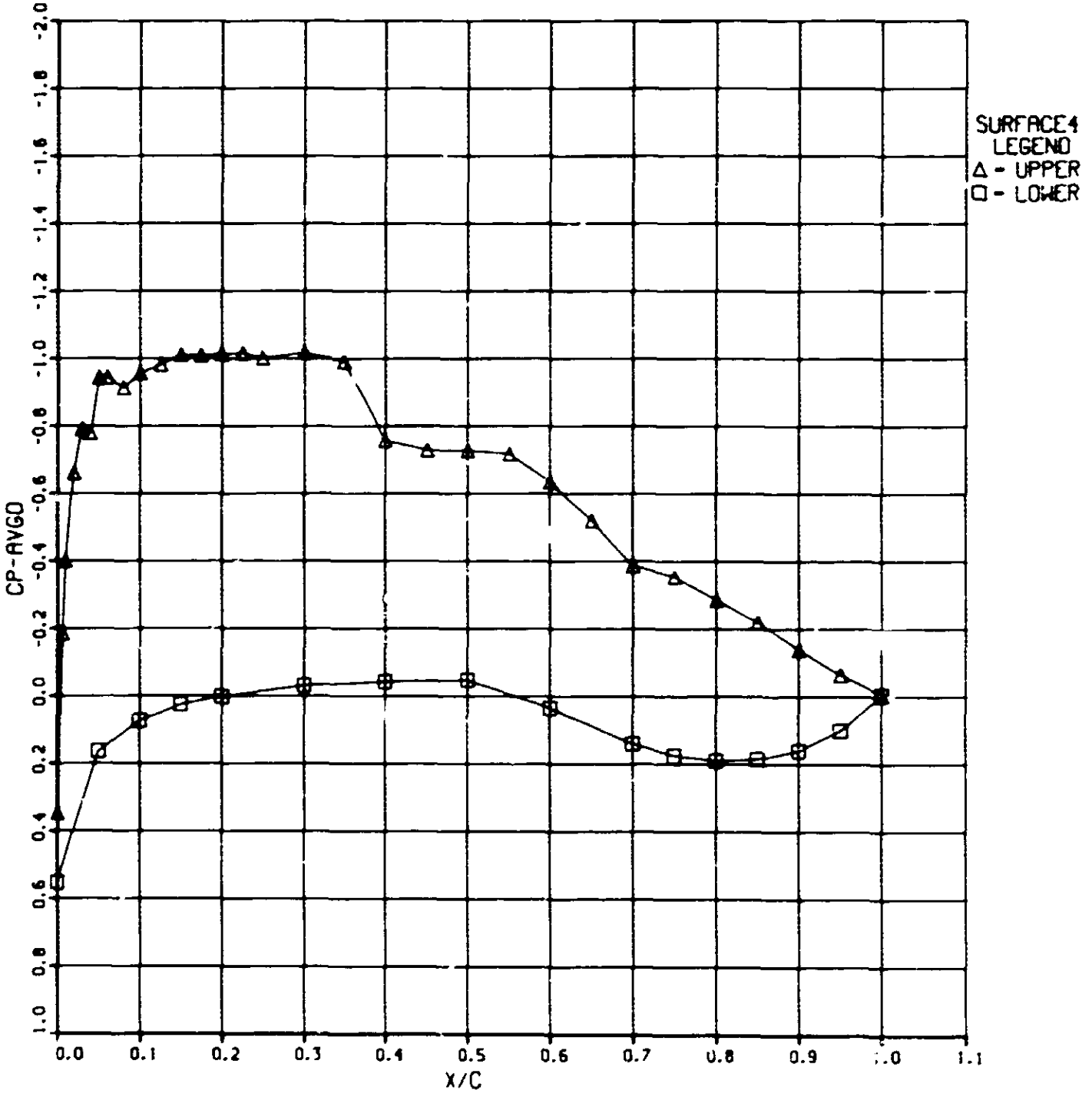


356-1-66 166.00: 3.00 CONF-17 MACH-0.893 RN-4.347 PT-2163 ALPHA- 5.00



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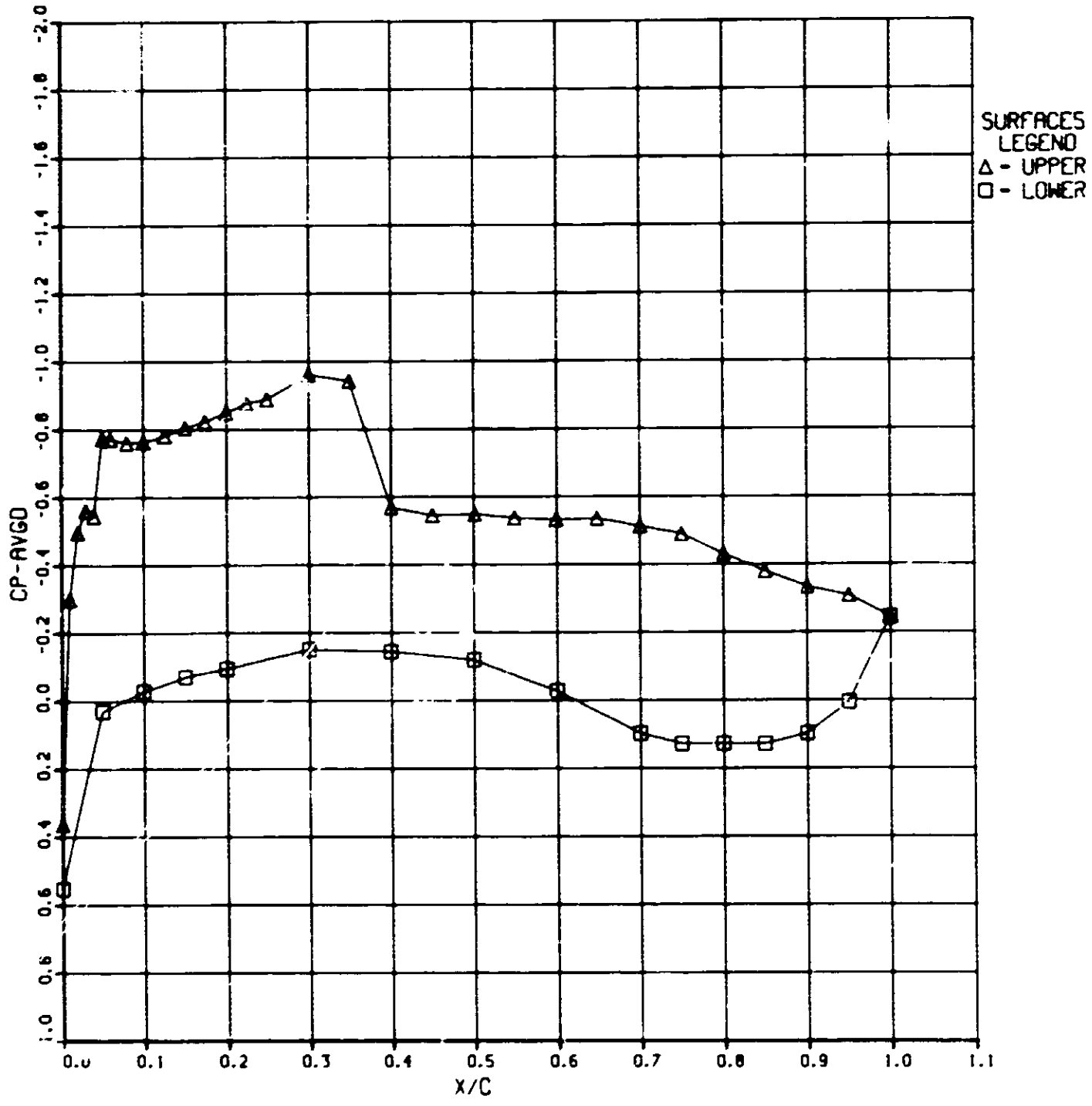
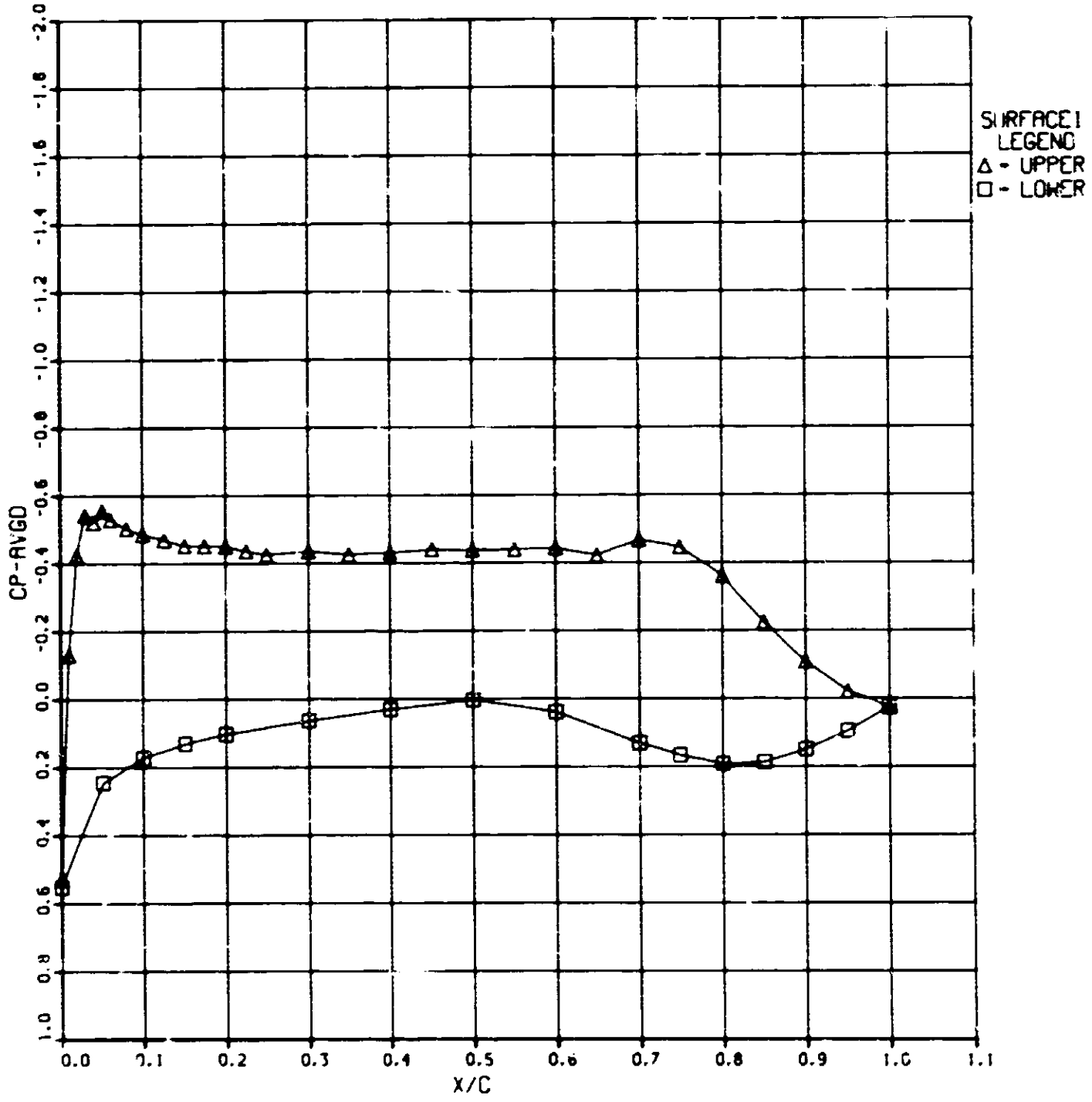


356-1-66 166.00: 3.00 CONF-17 MACH-0.893 RN-4.347 PT-2163 ALPHA- 5.00



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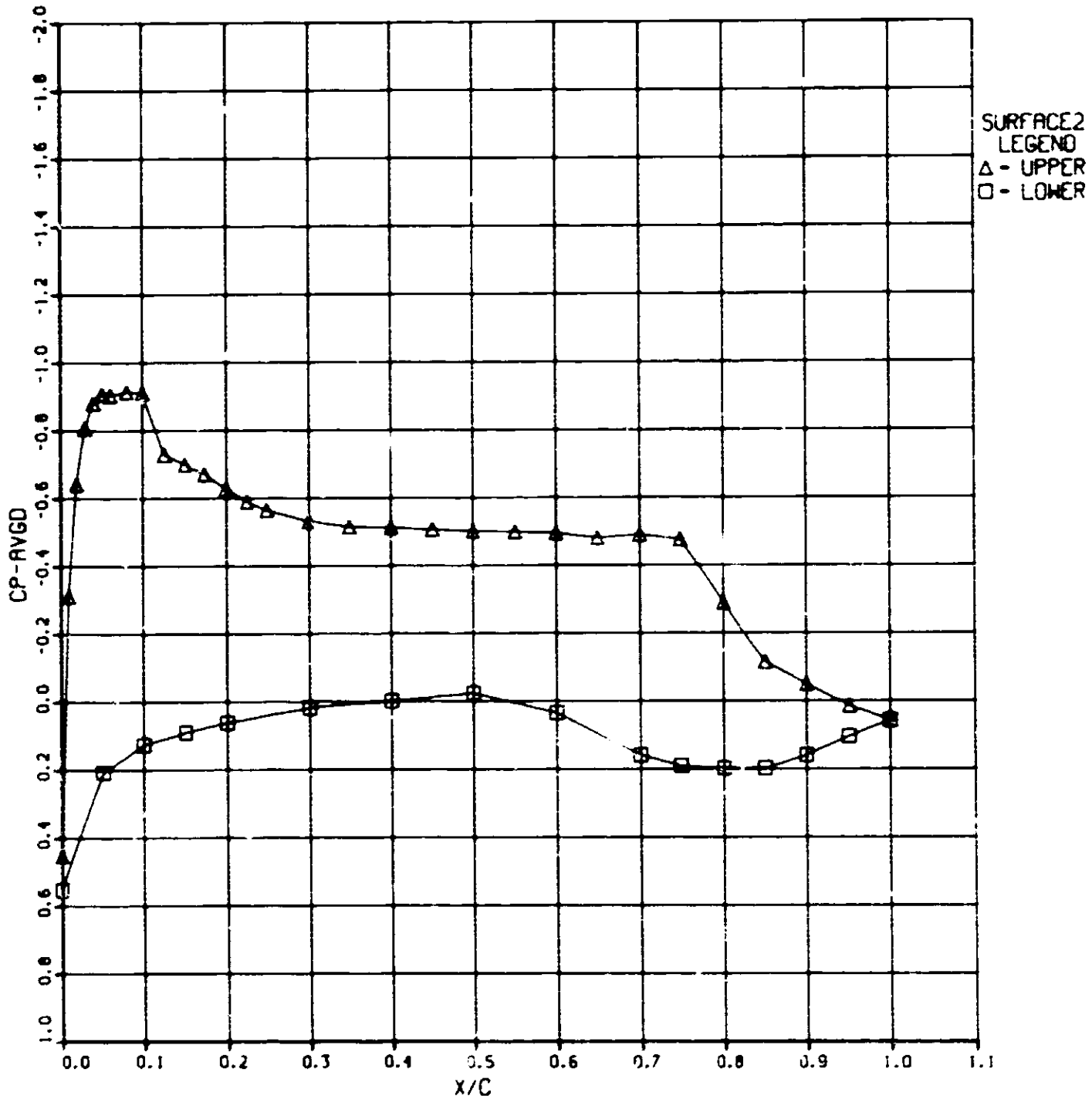

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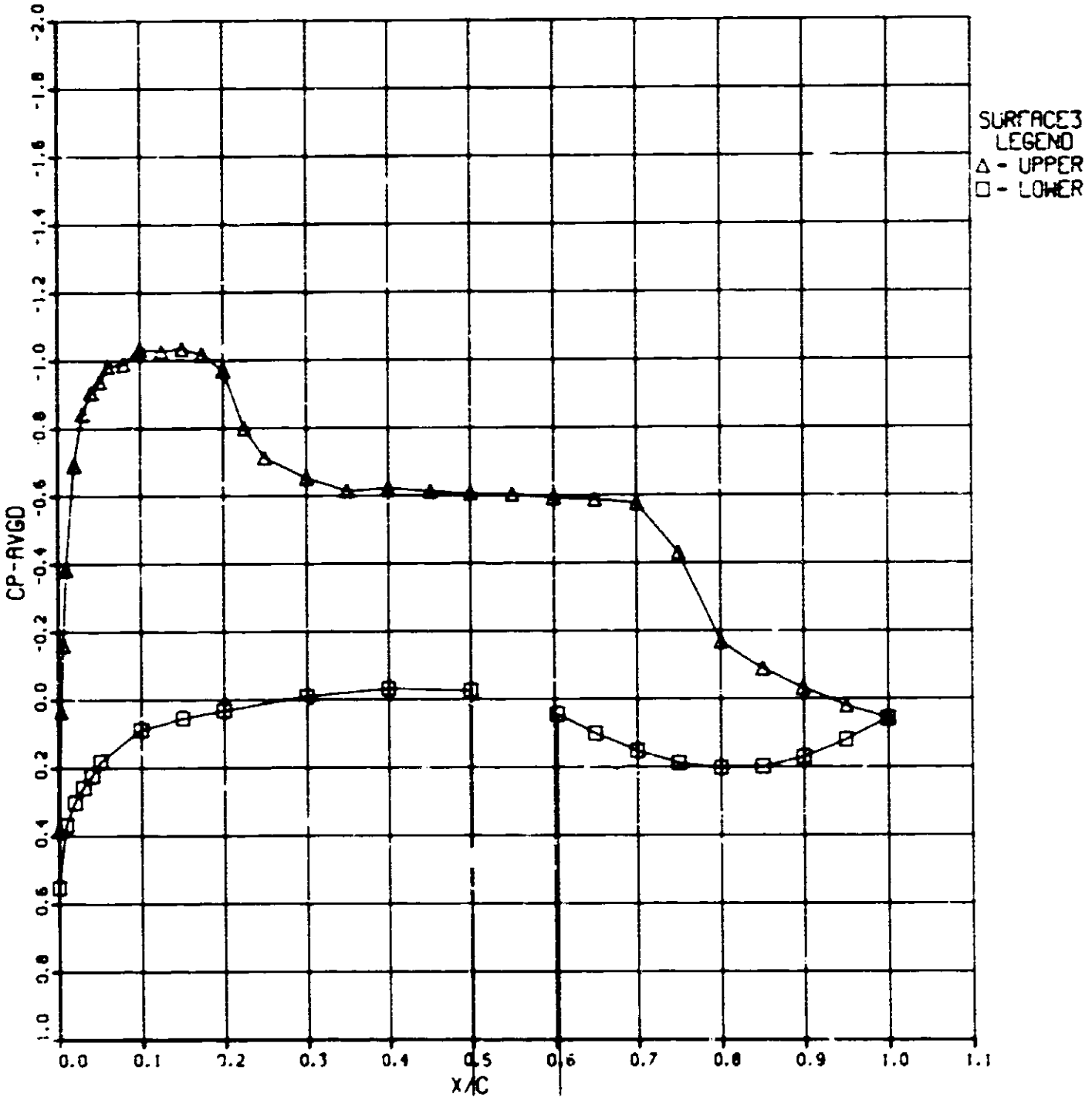

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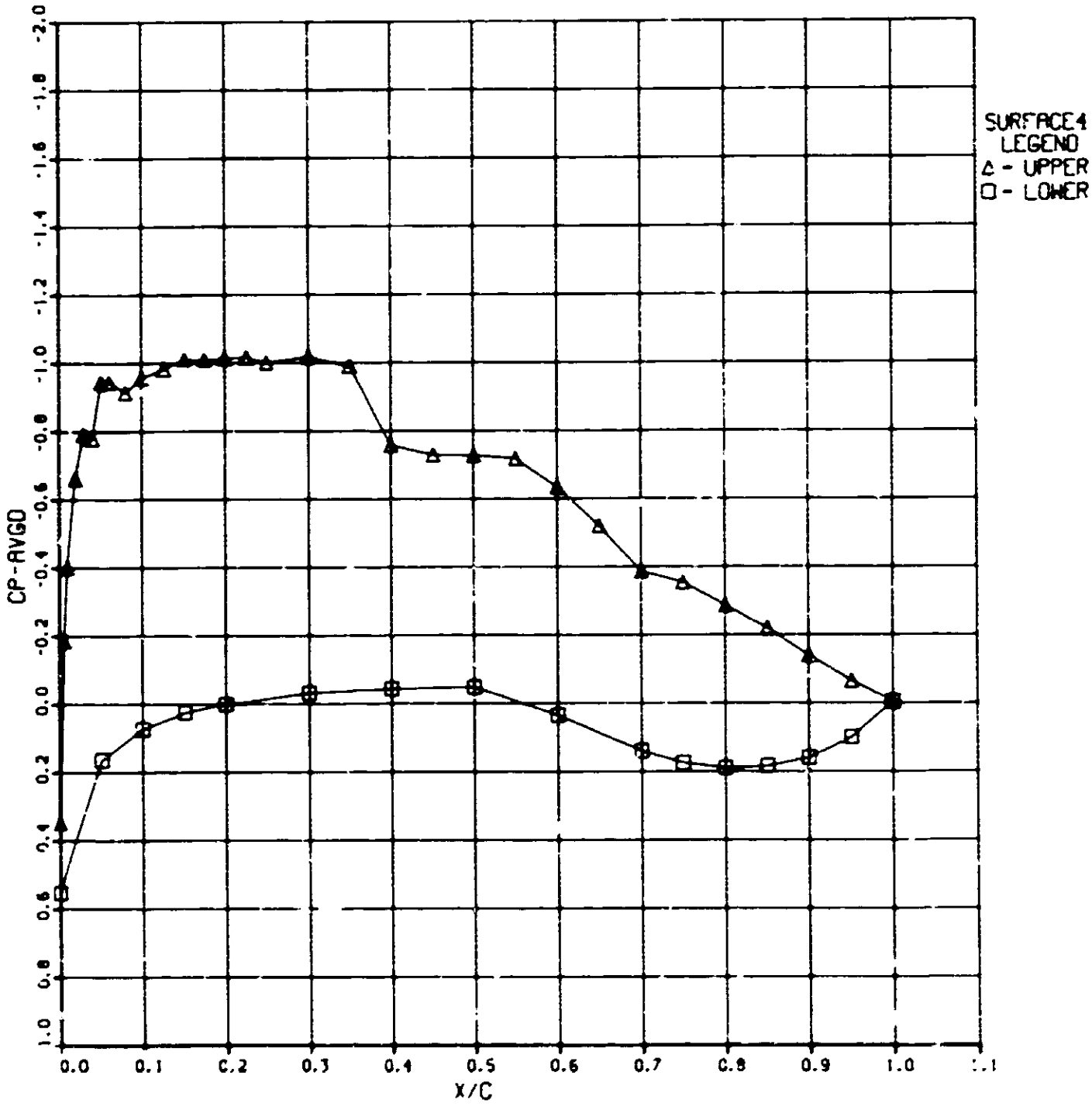

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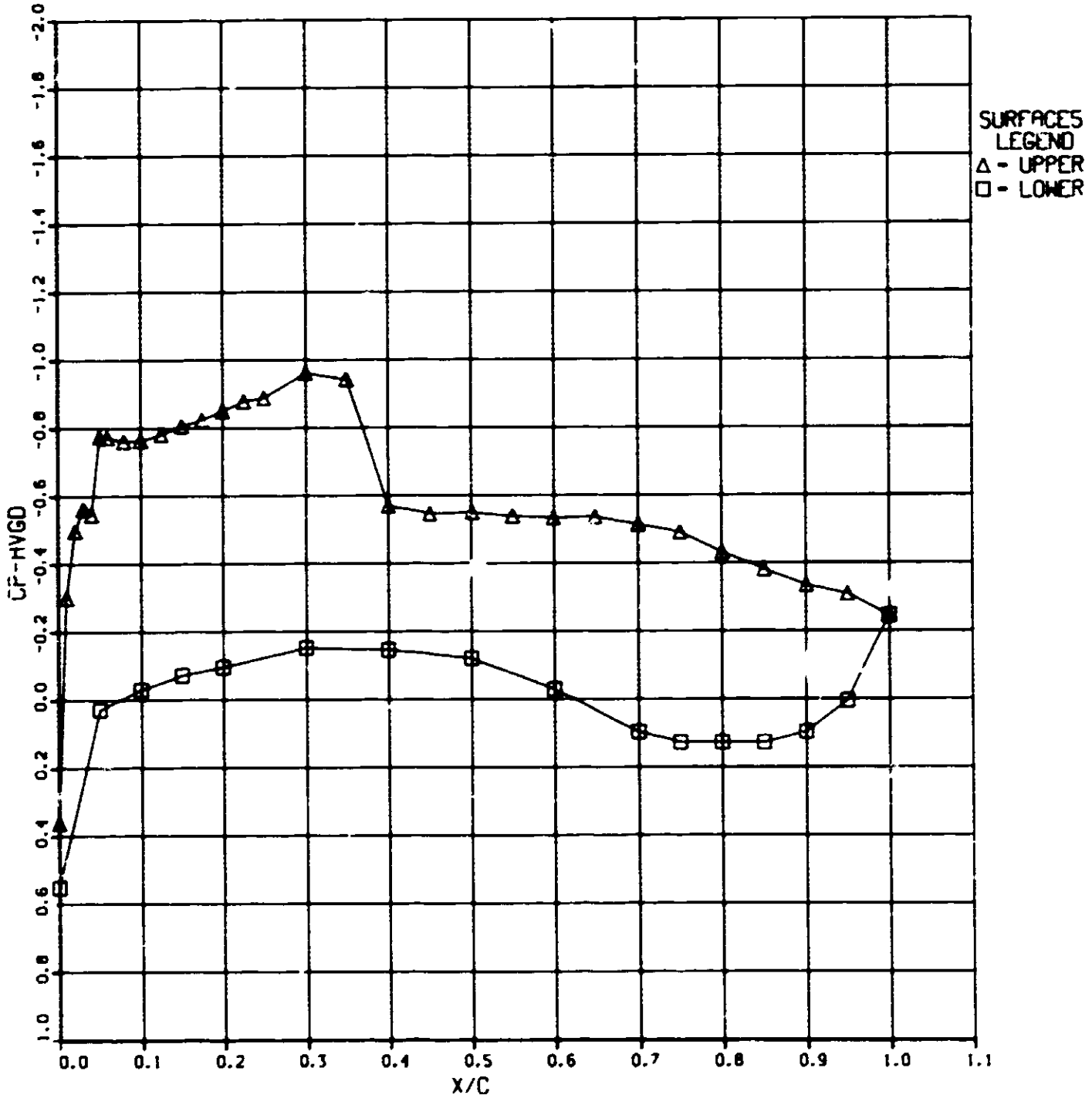
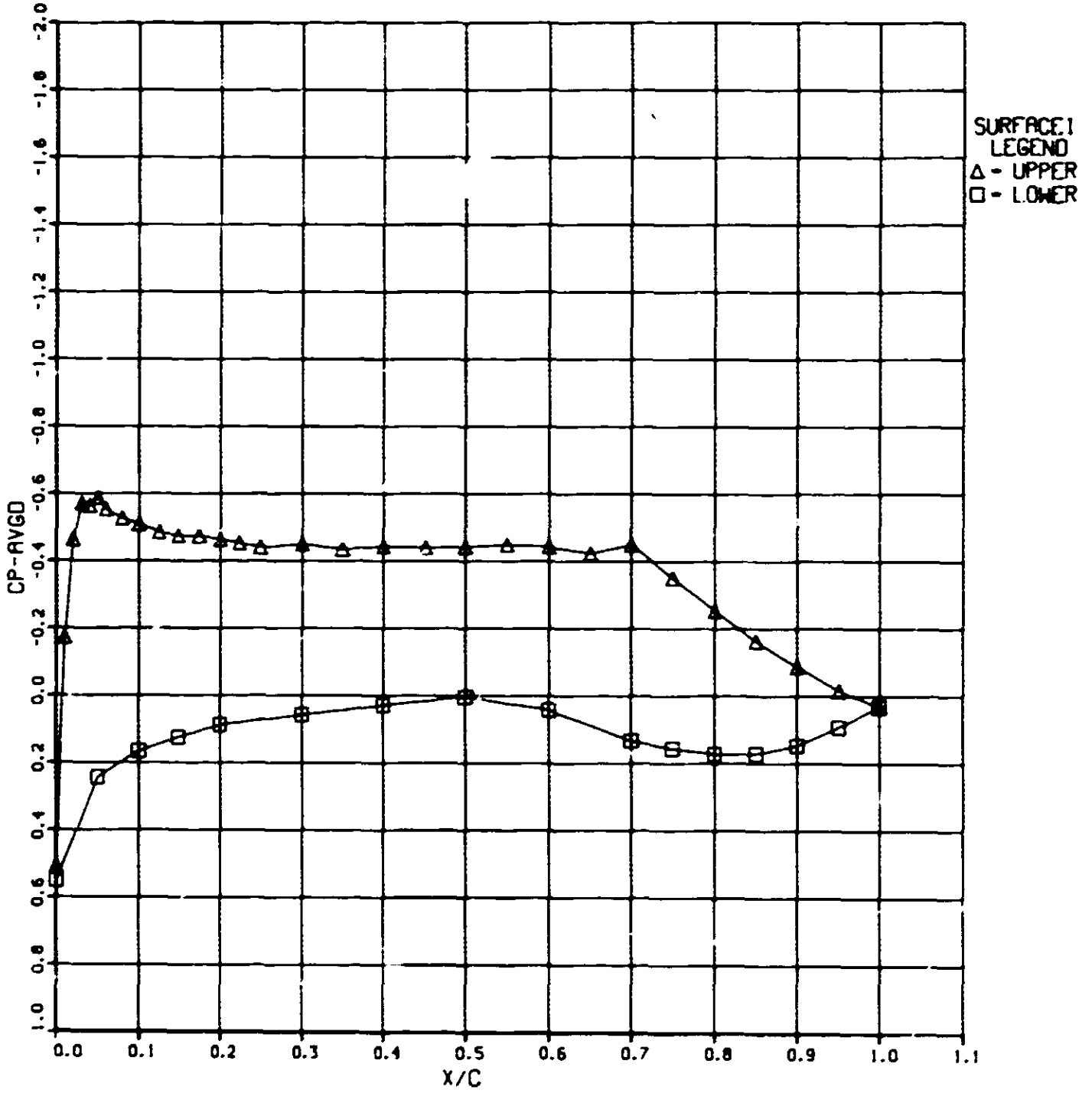


356-1-66 167.00: 2.00 CONF-17 MACH-0.877 RN-4.386 PT-2211 ALPHA- 5.00



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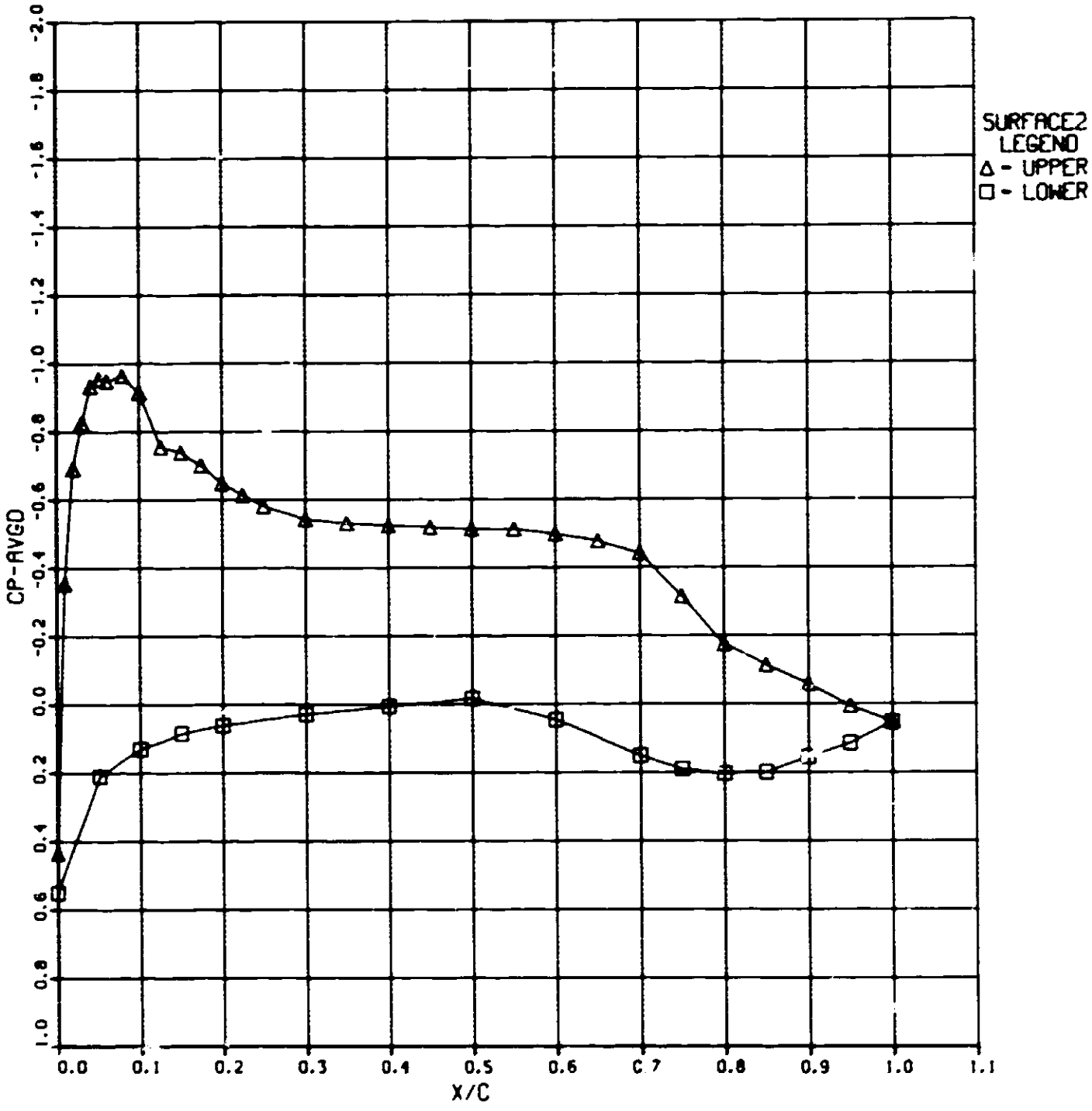

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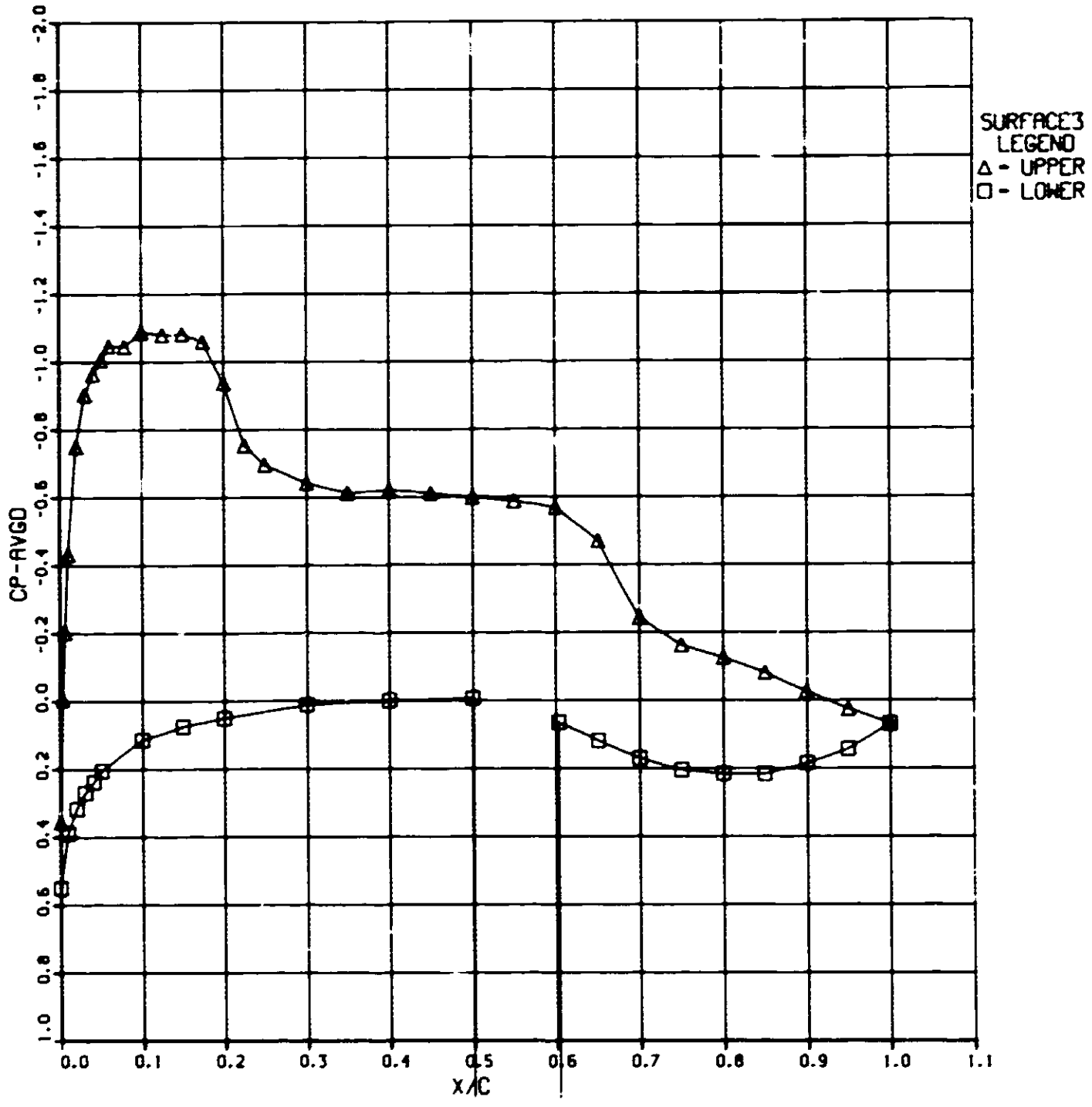

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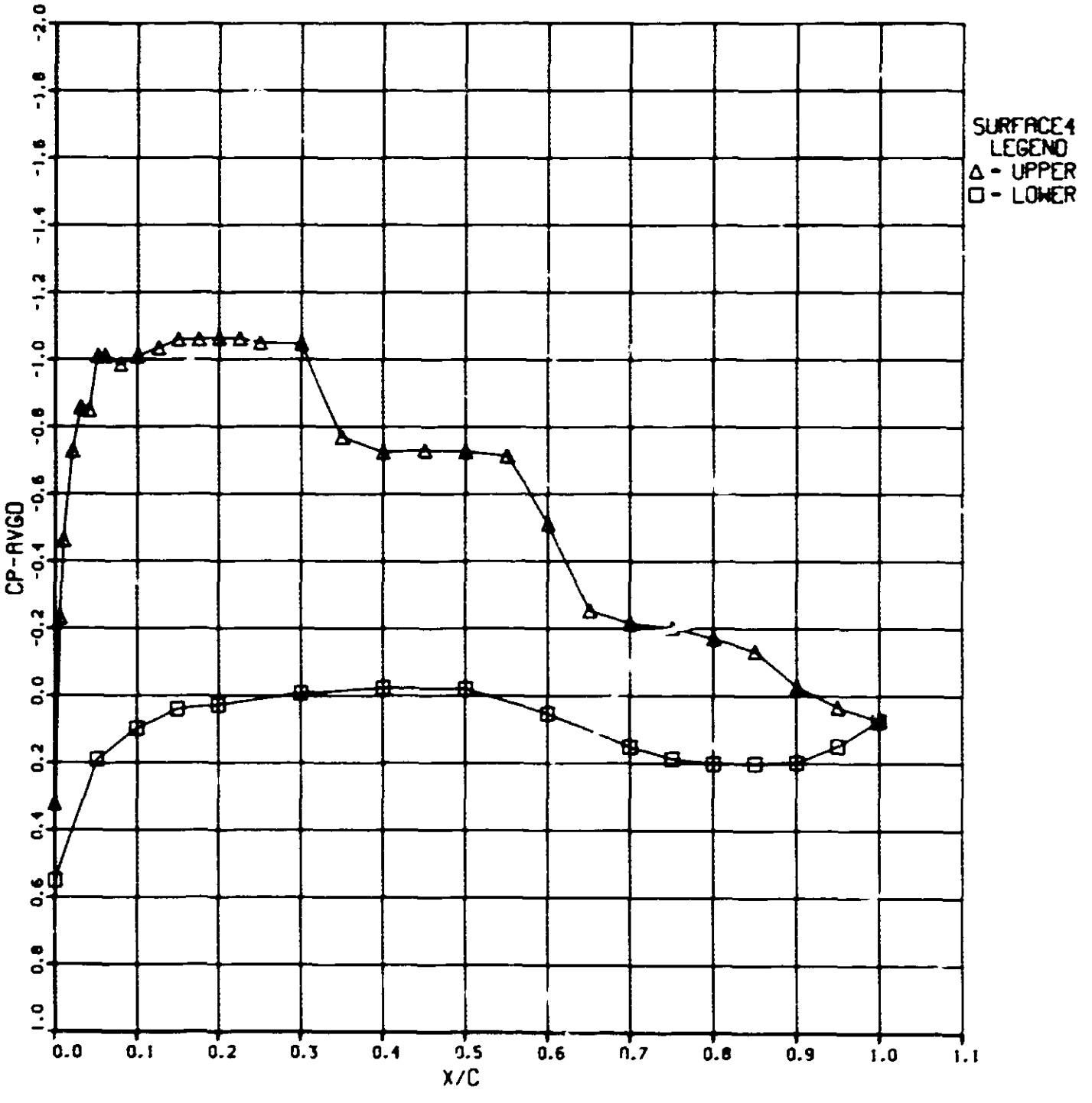

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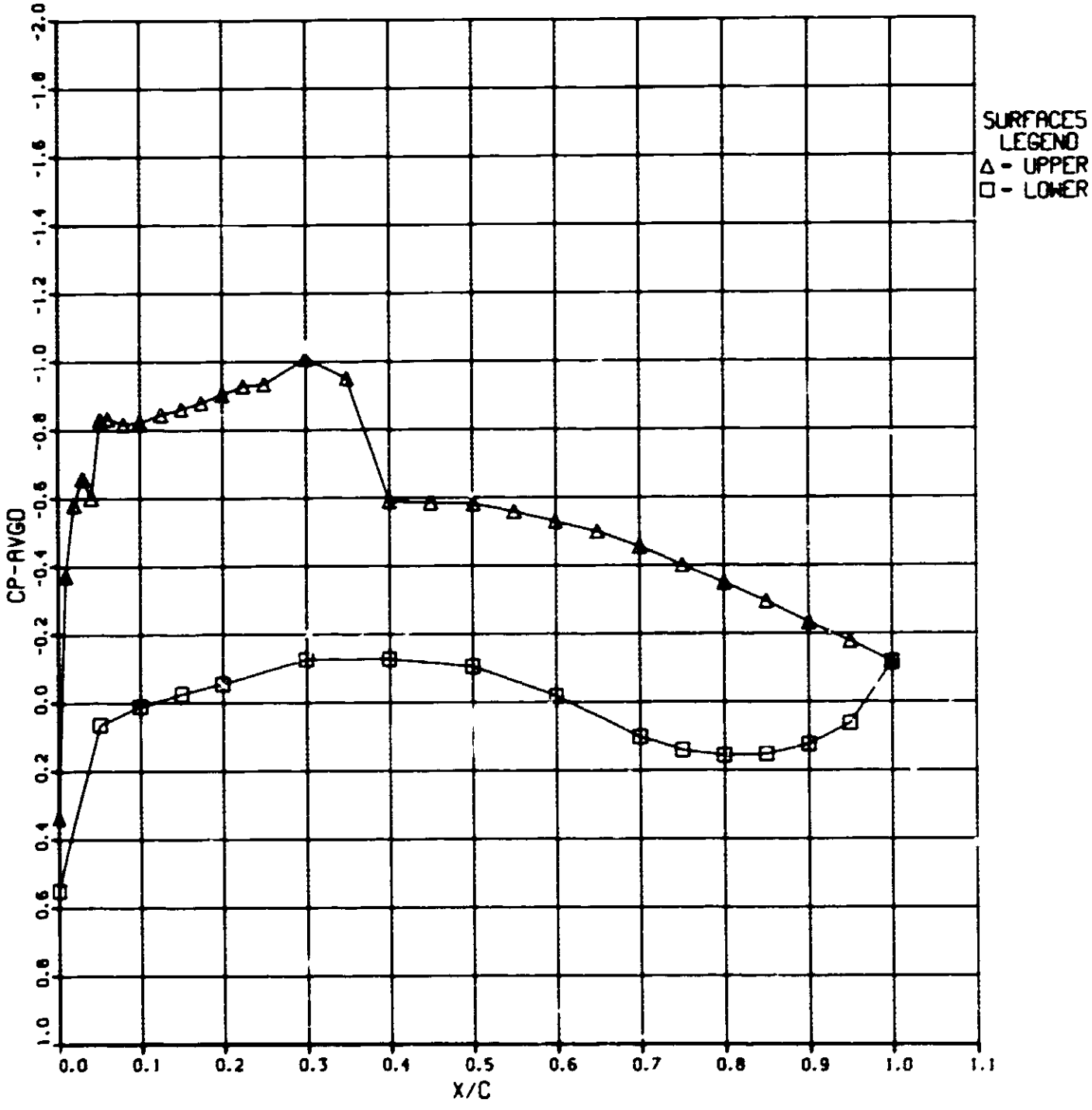

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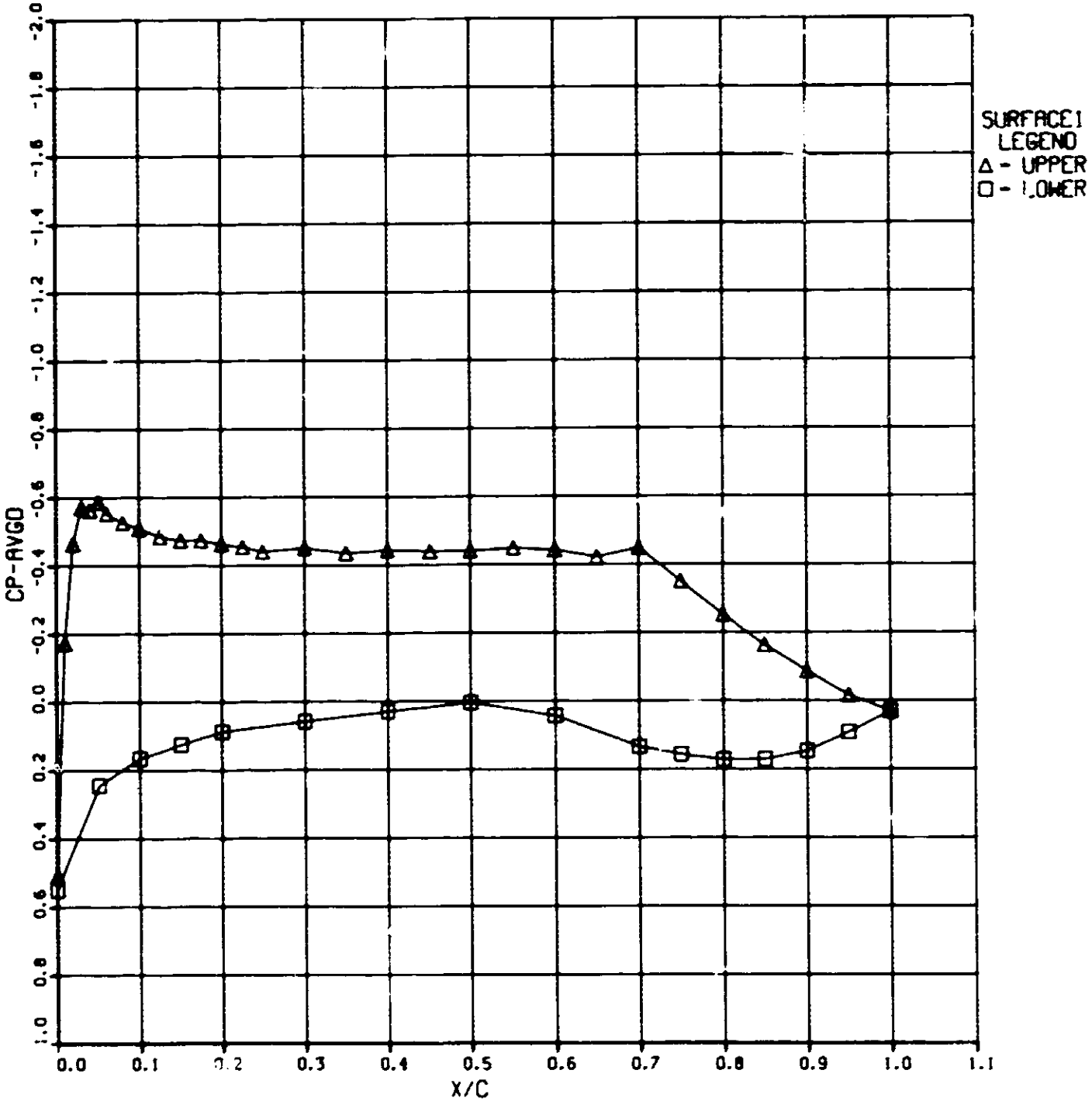

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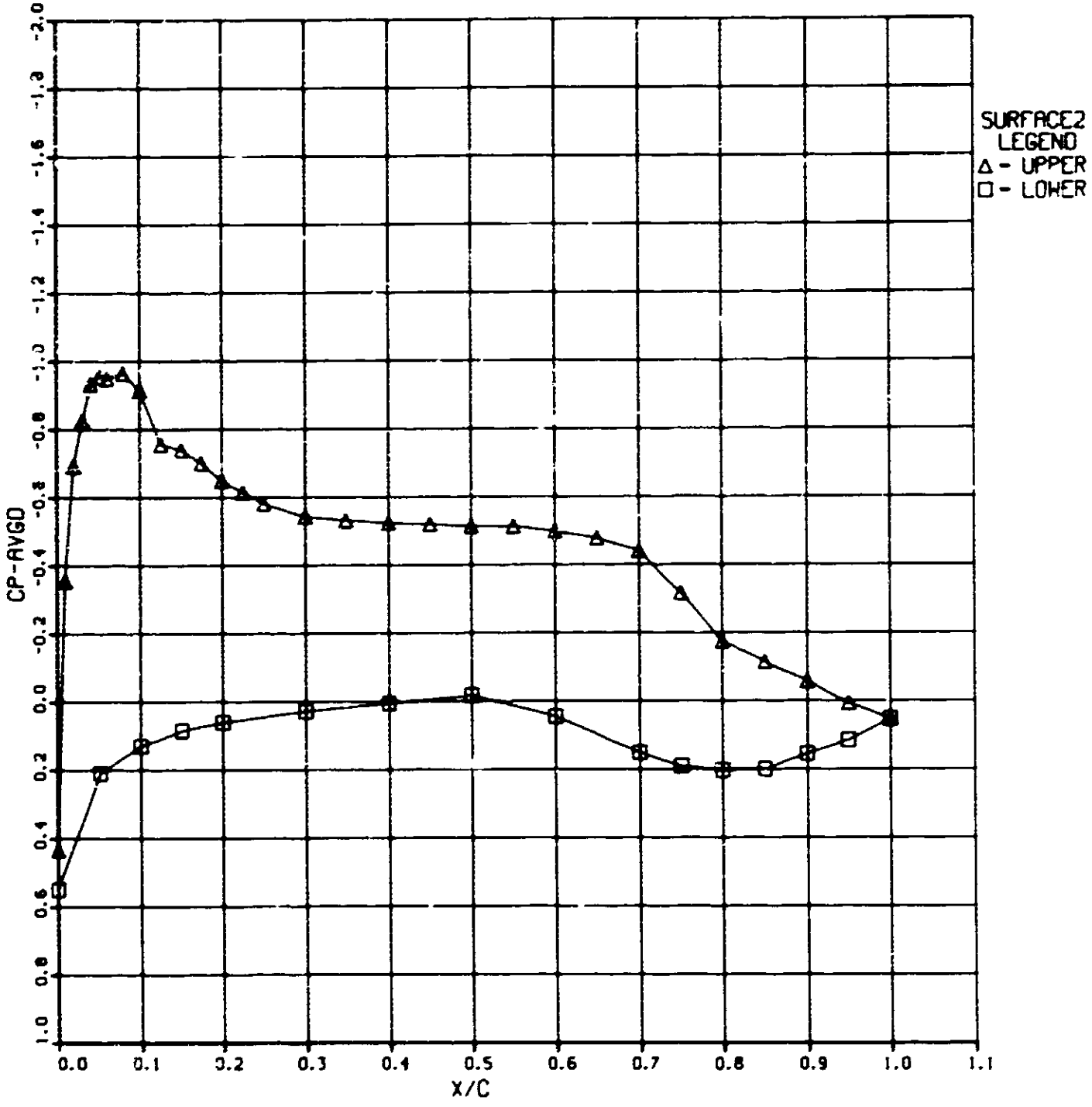

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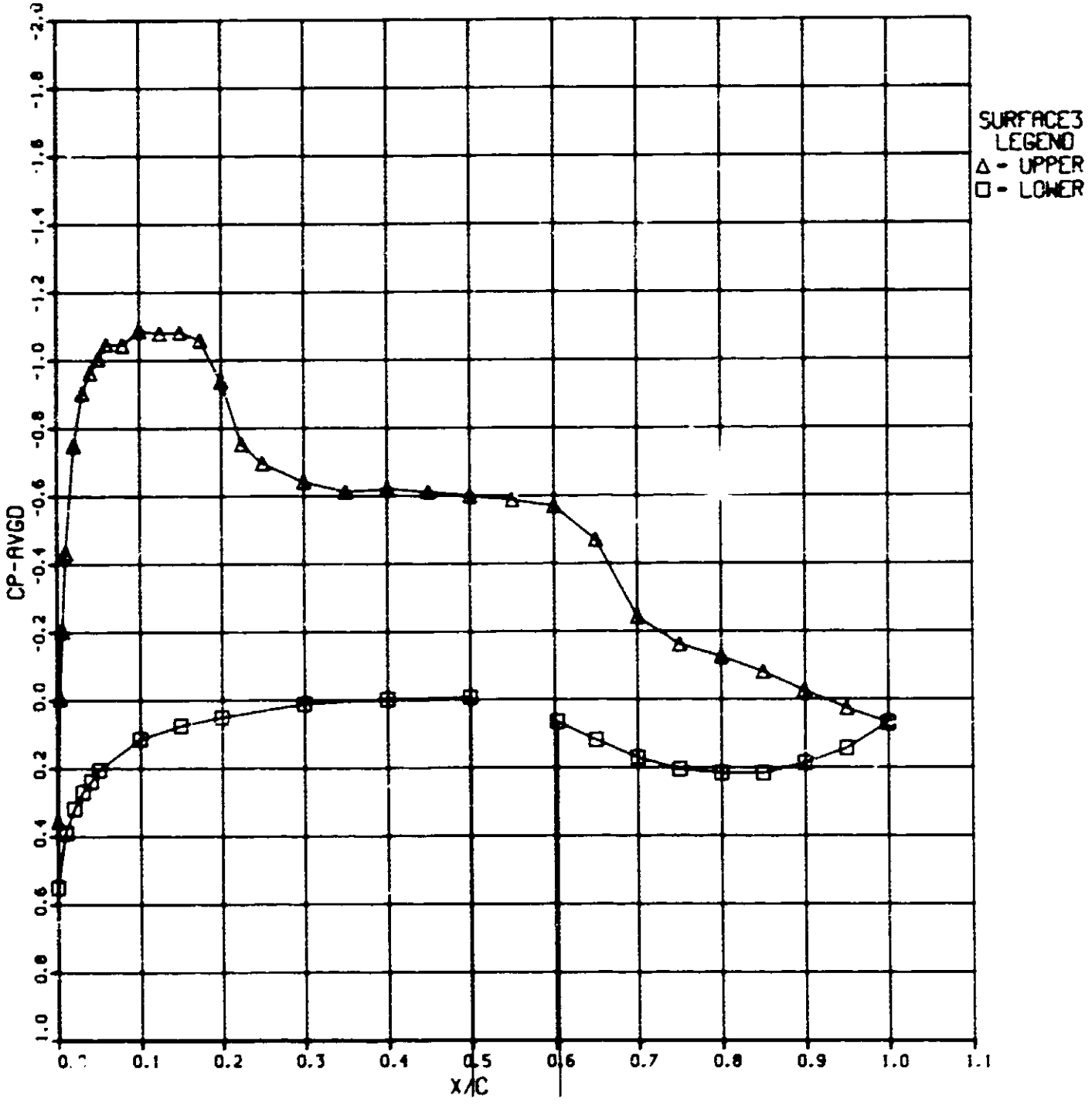

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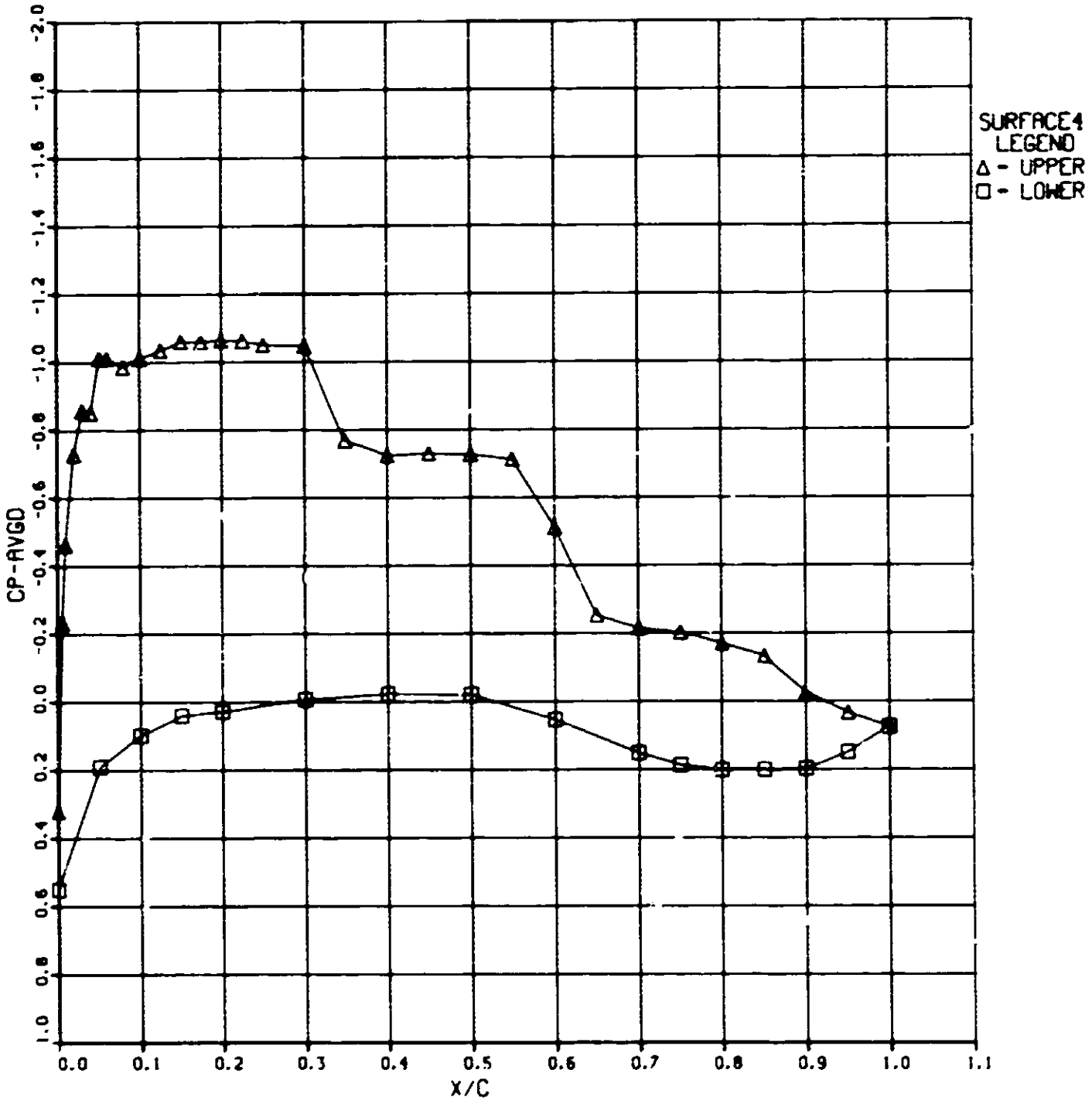

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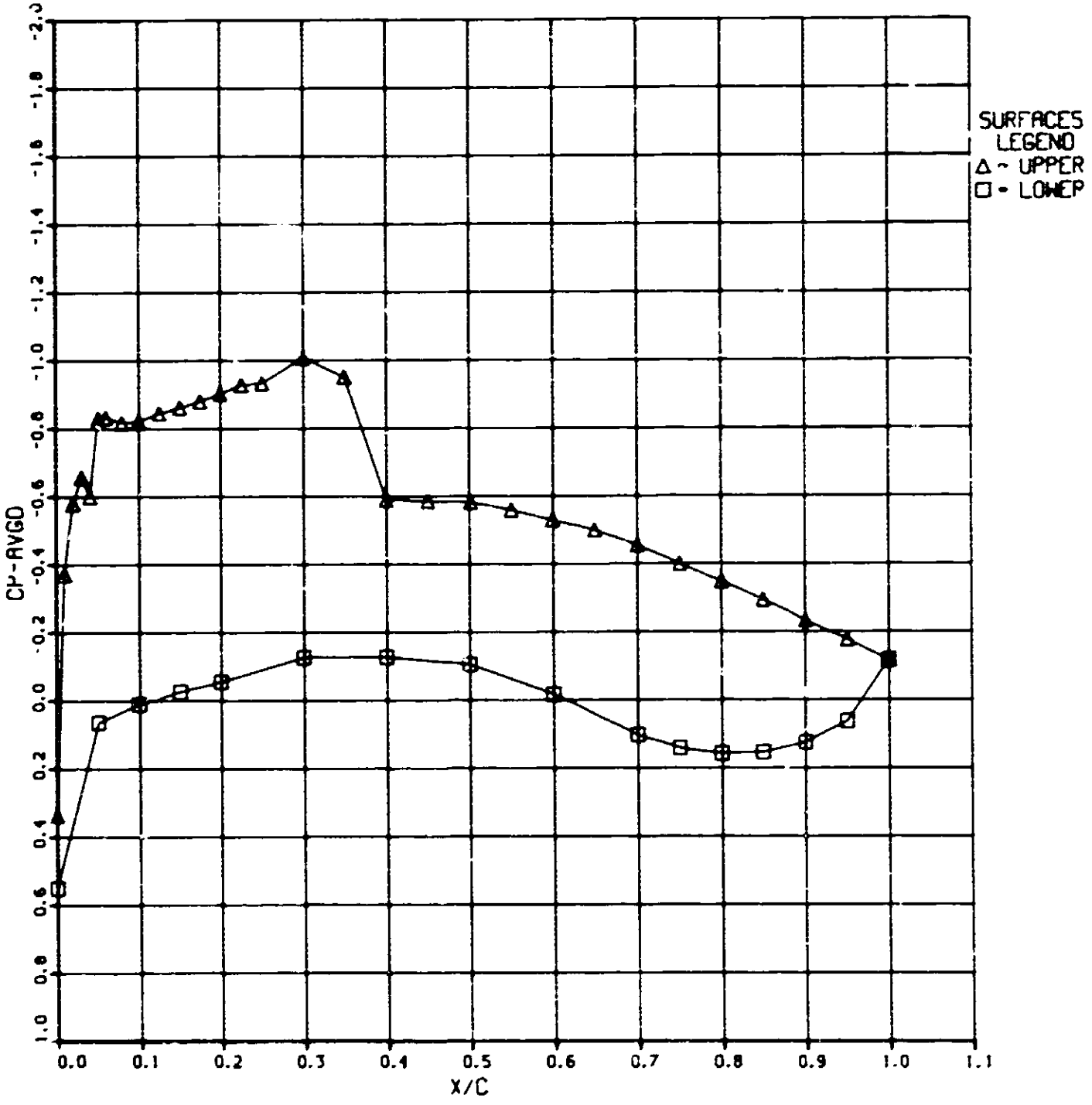

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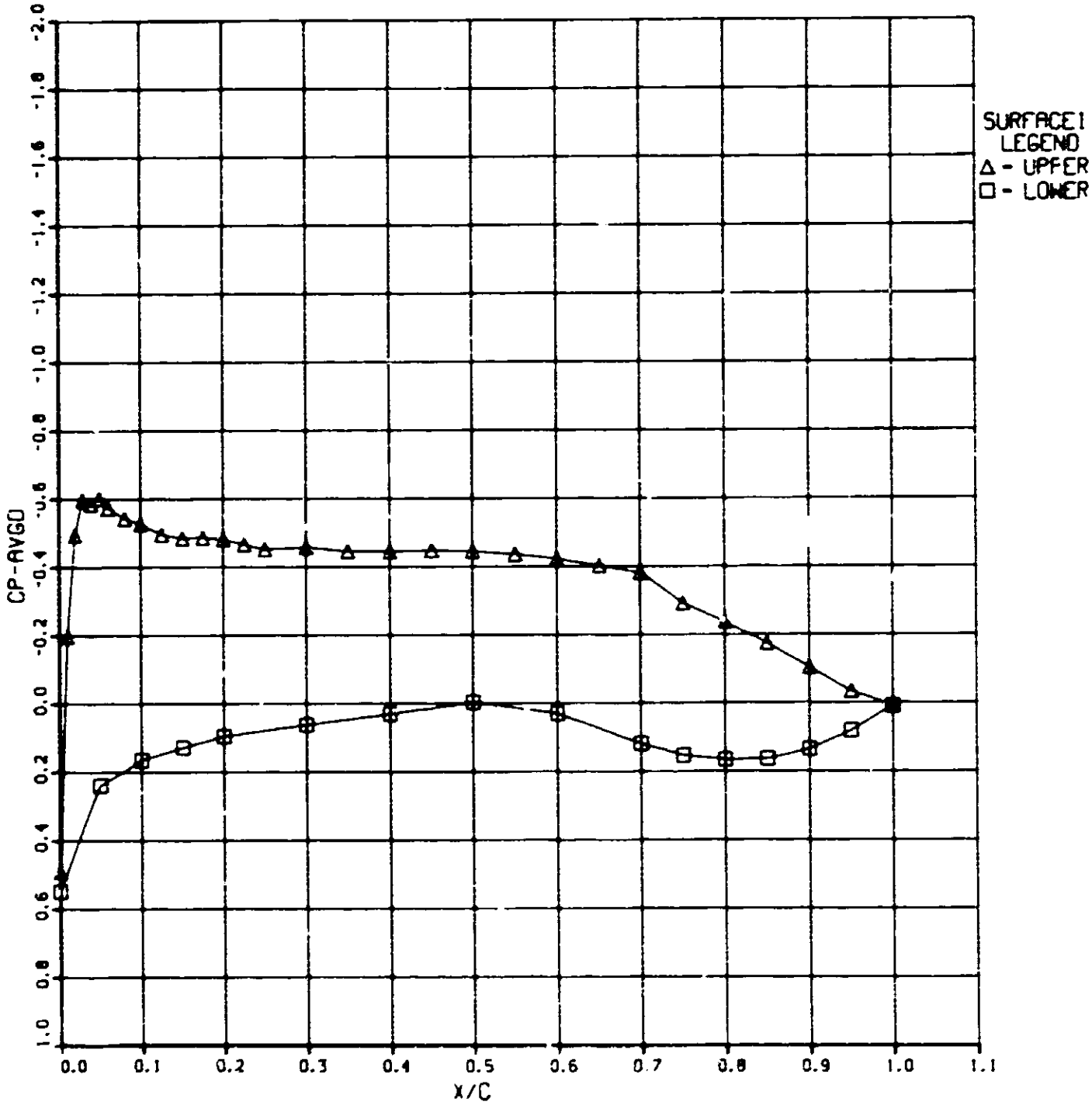

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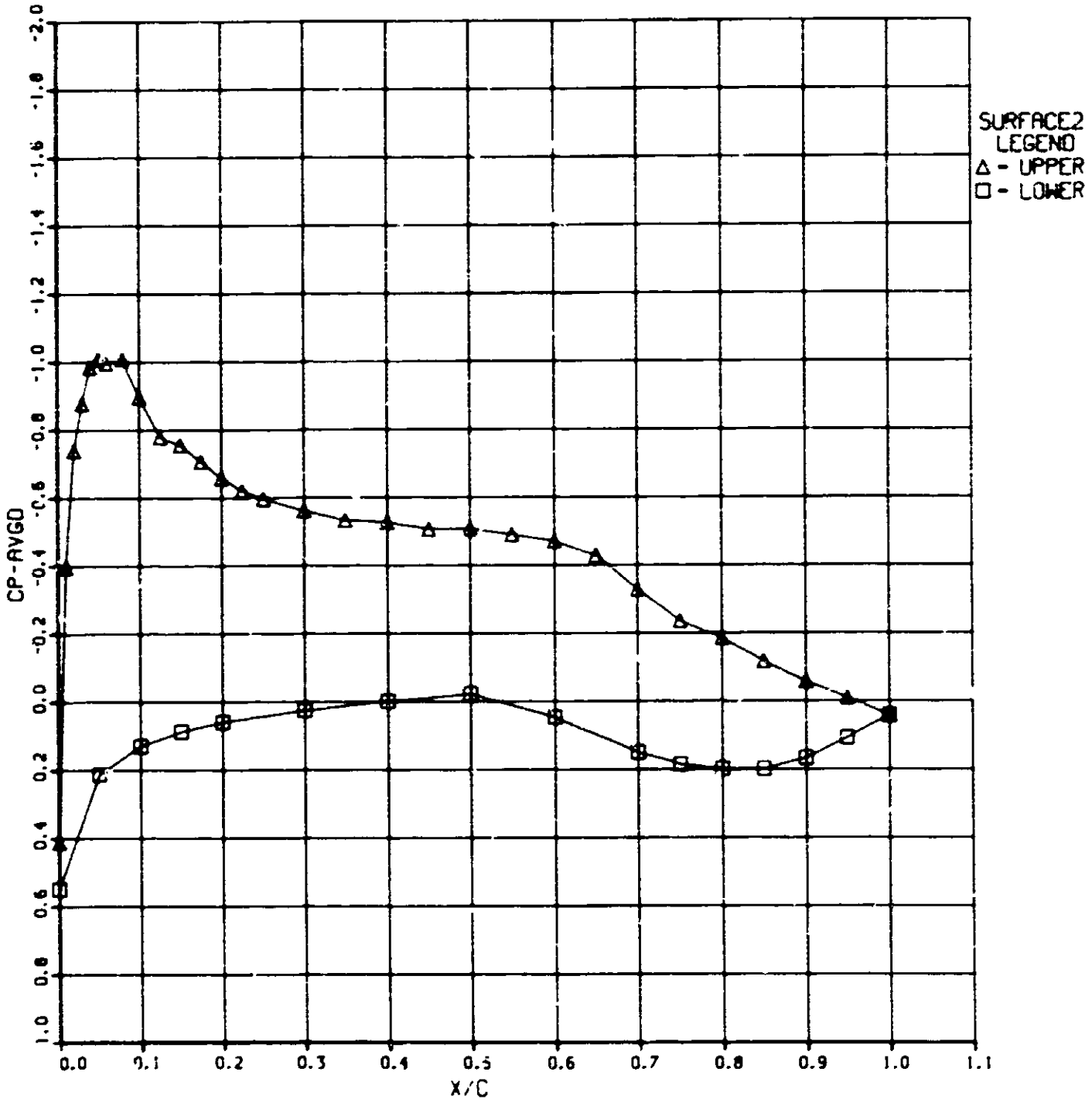

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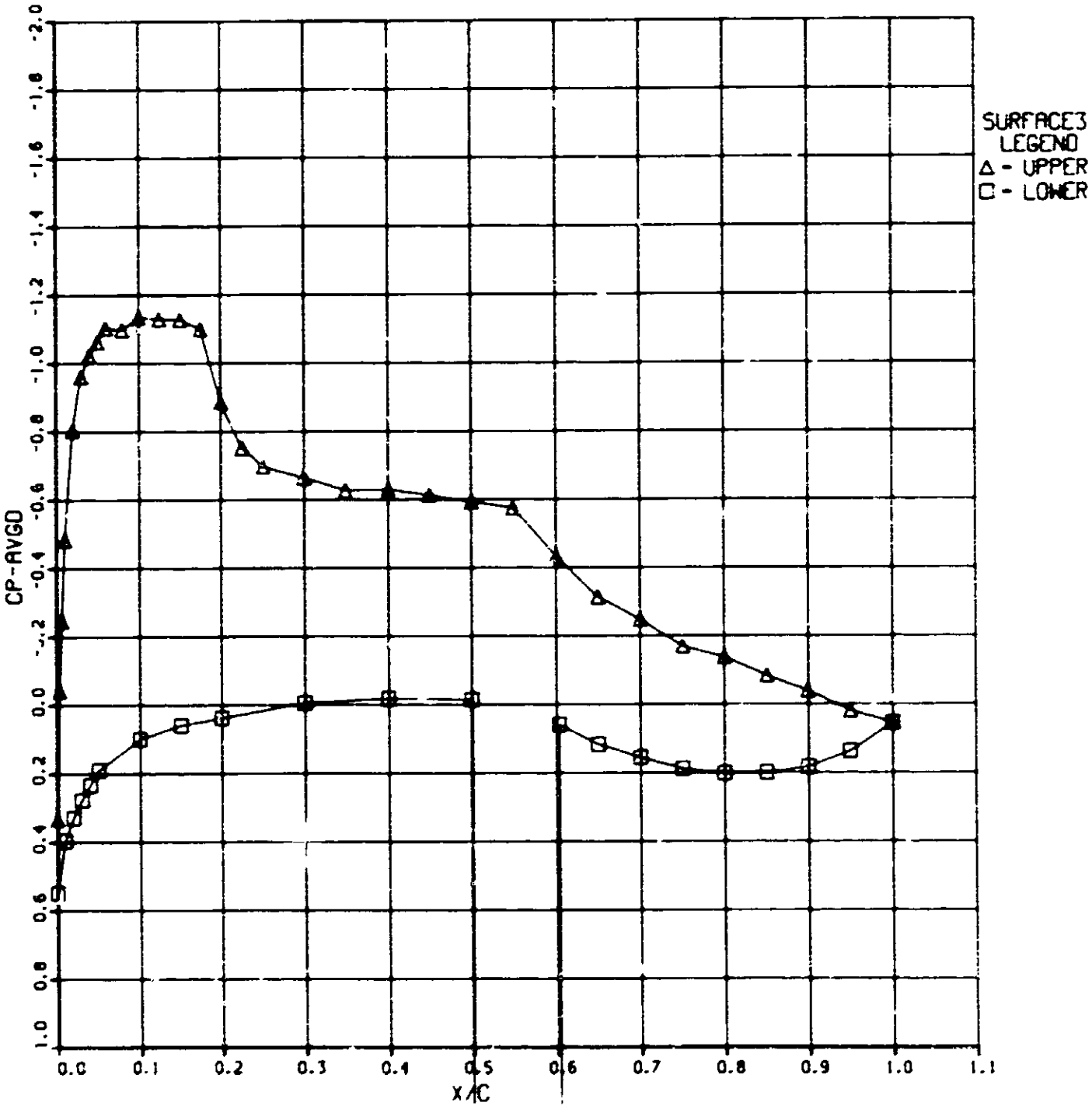

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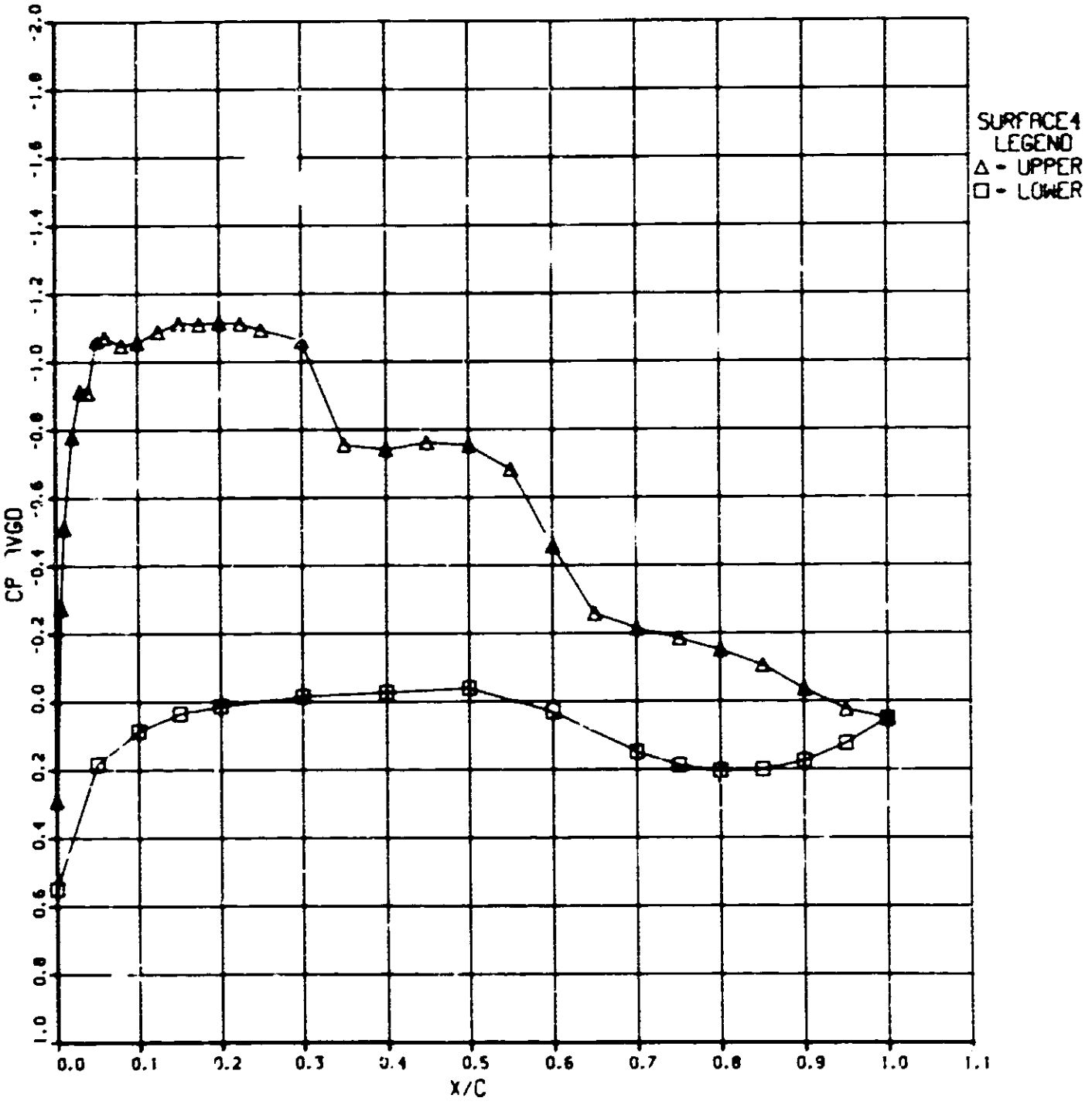

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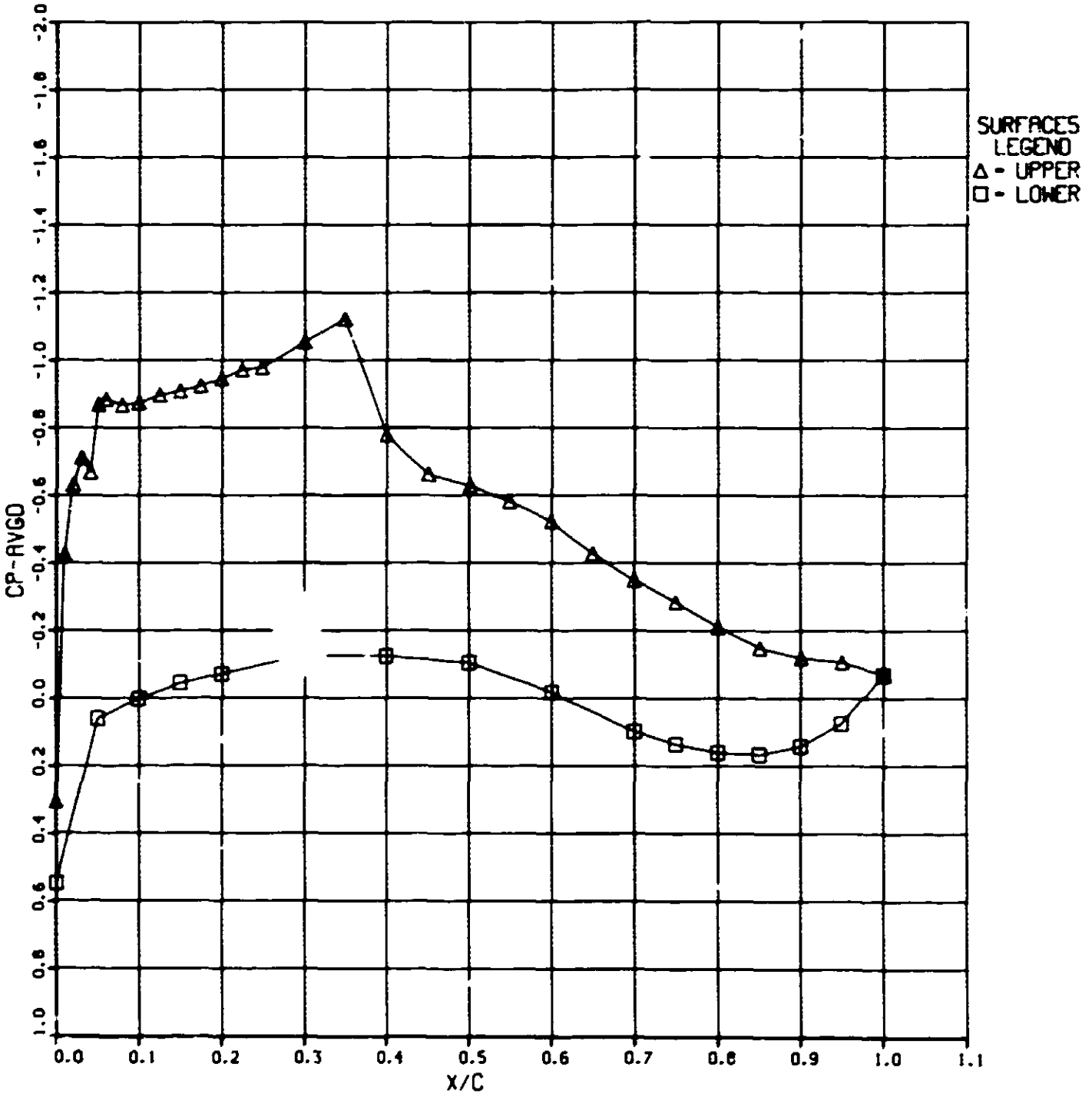

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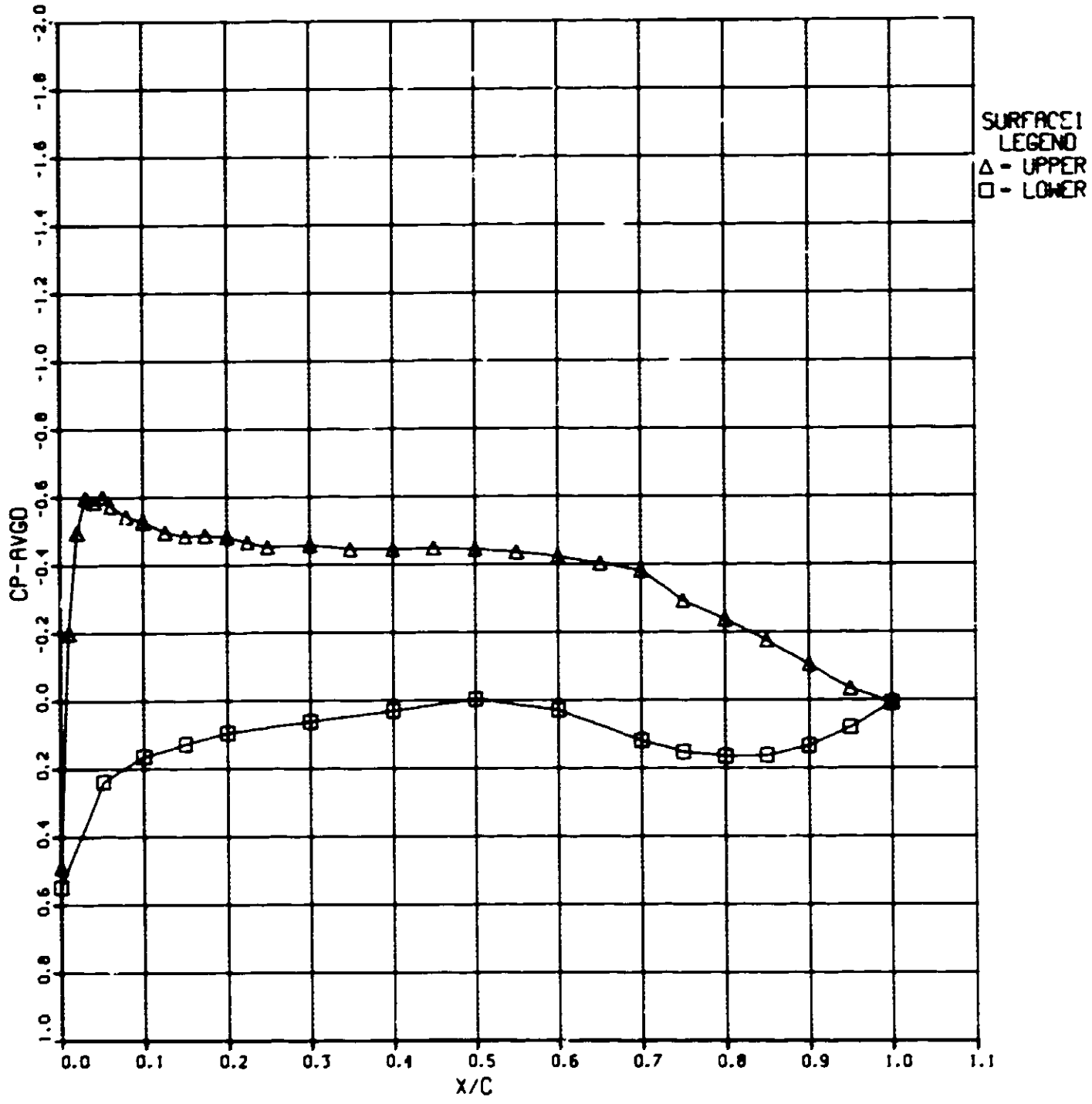

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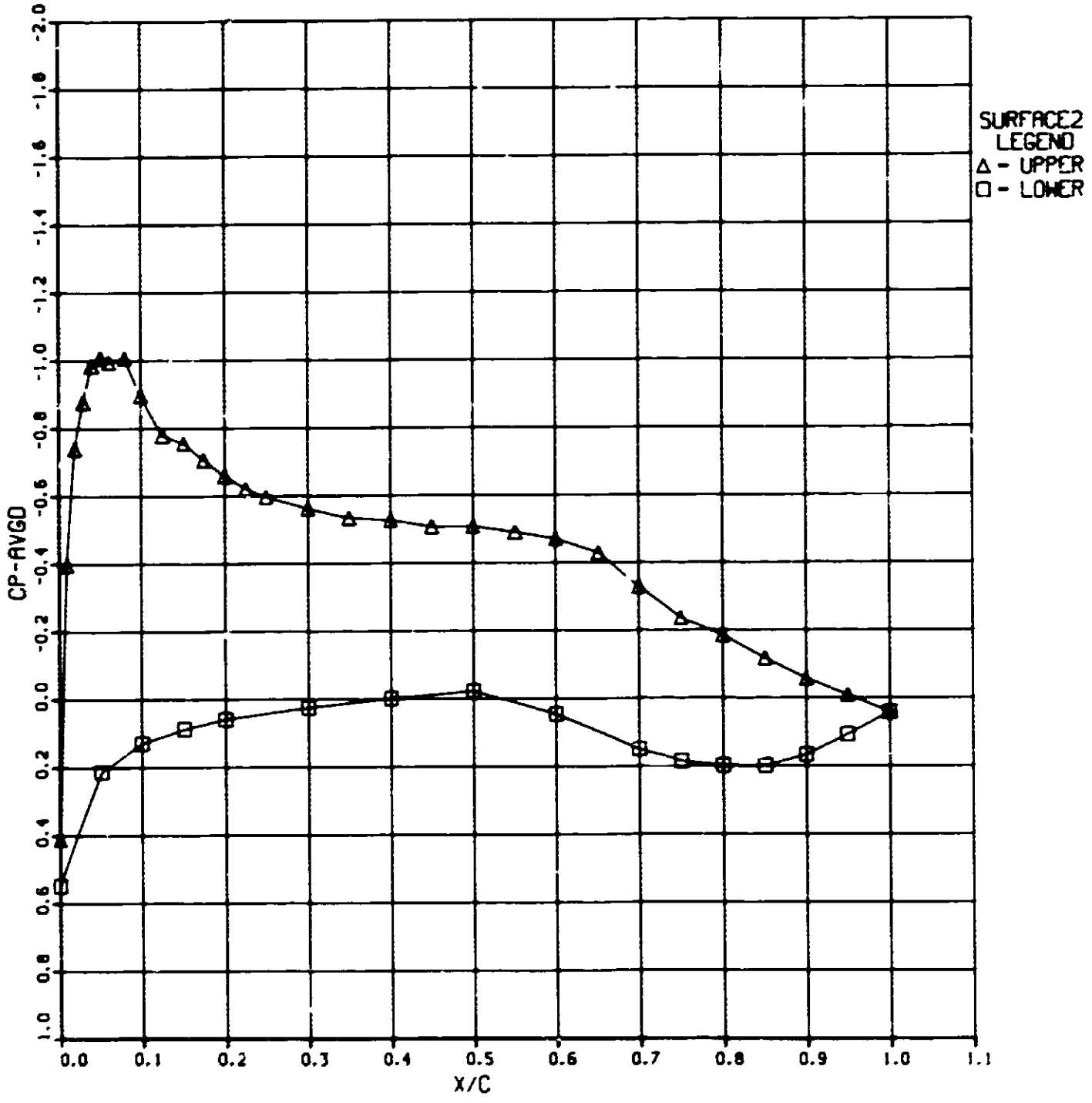

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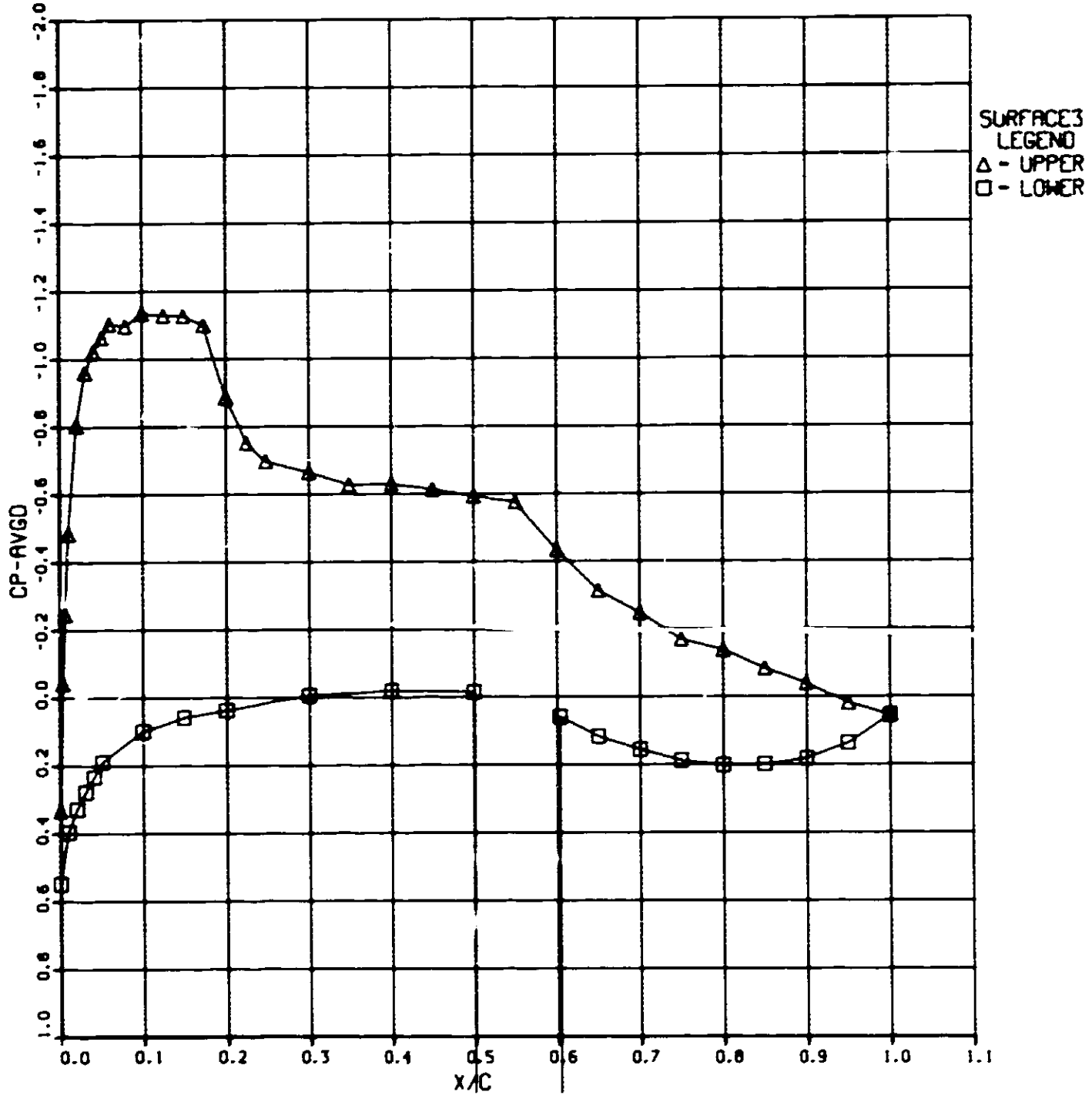

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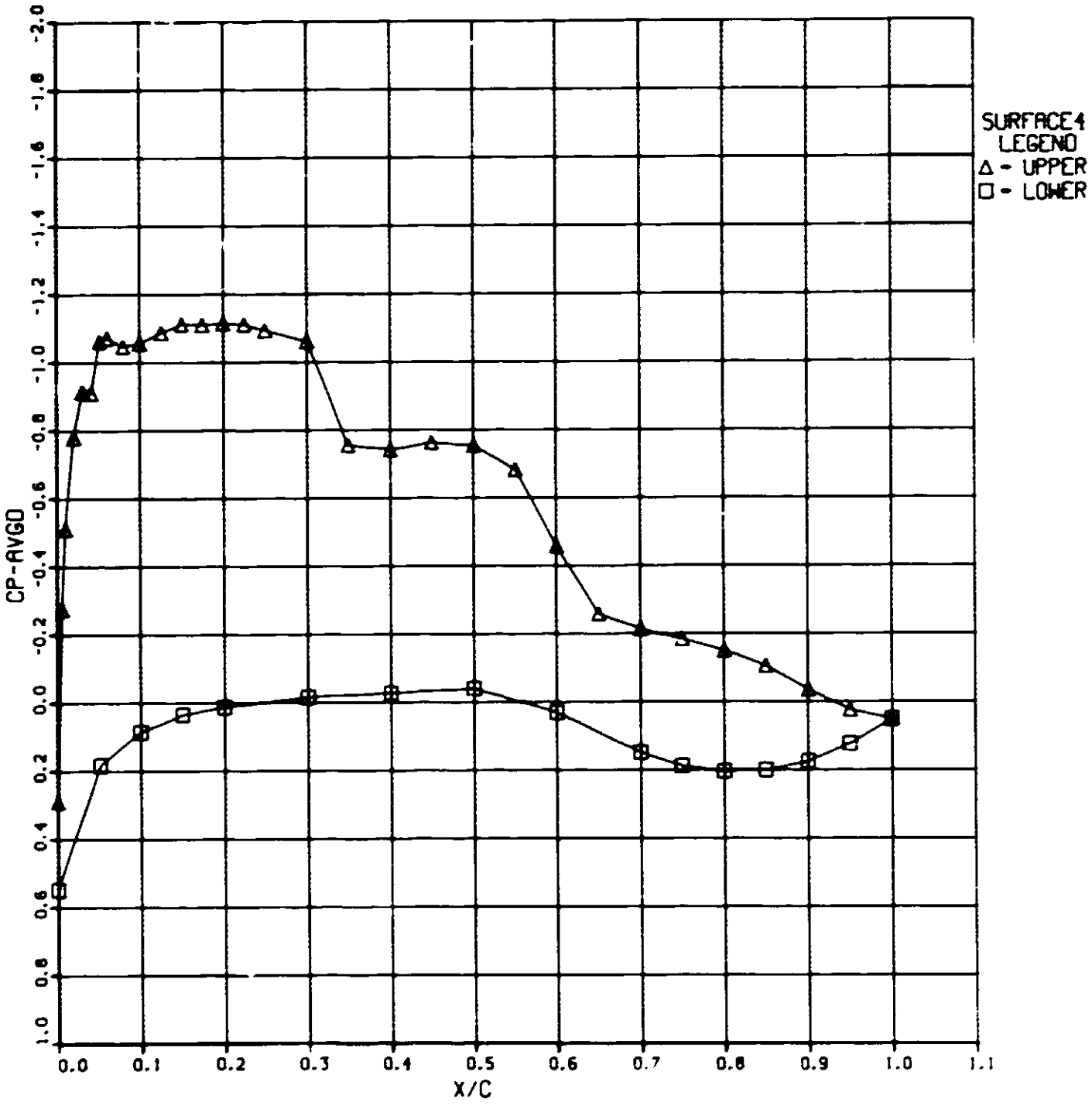

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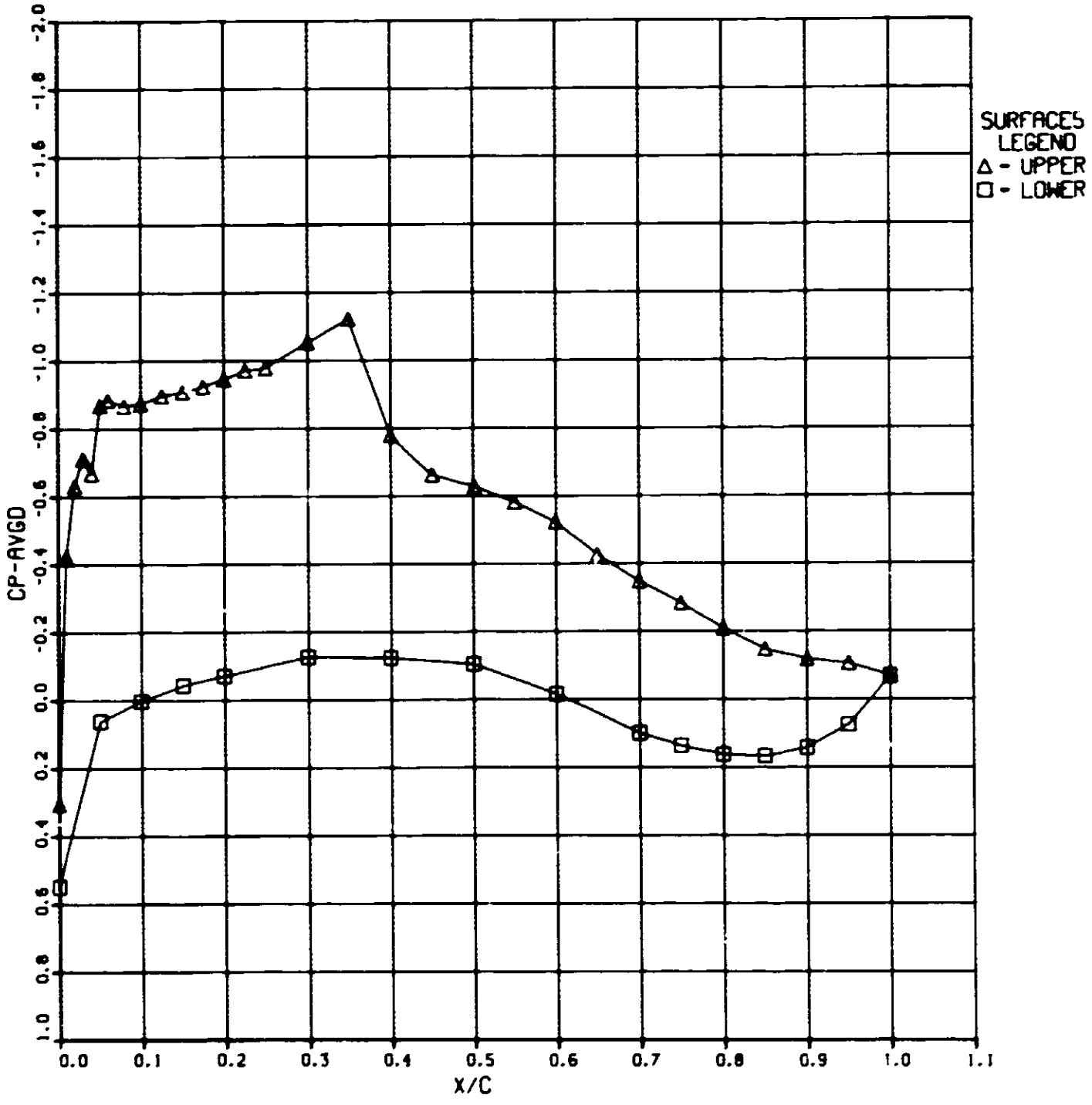
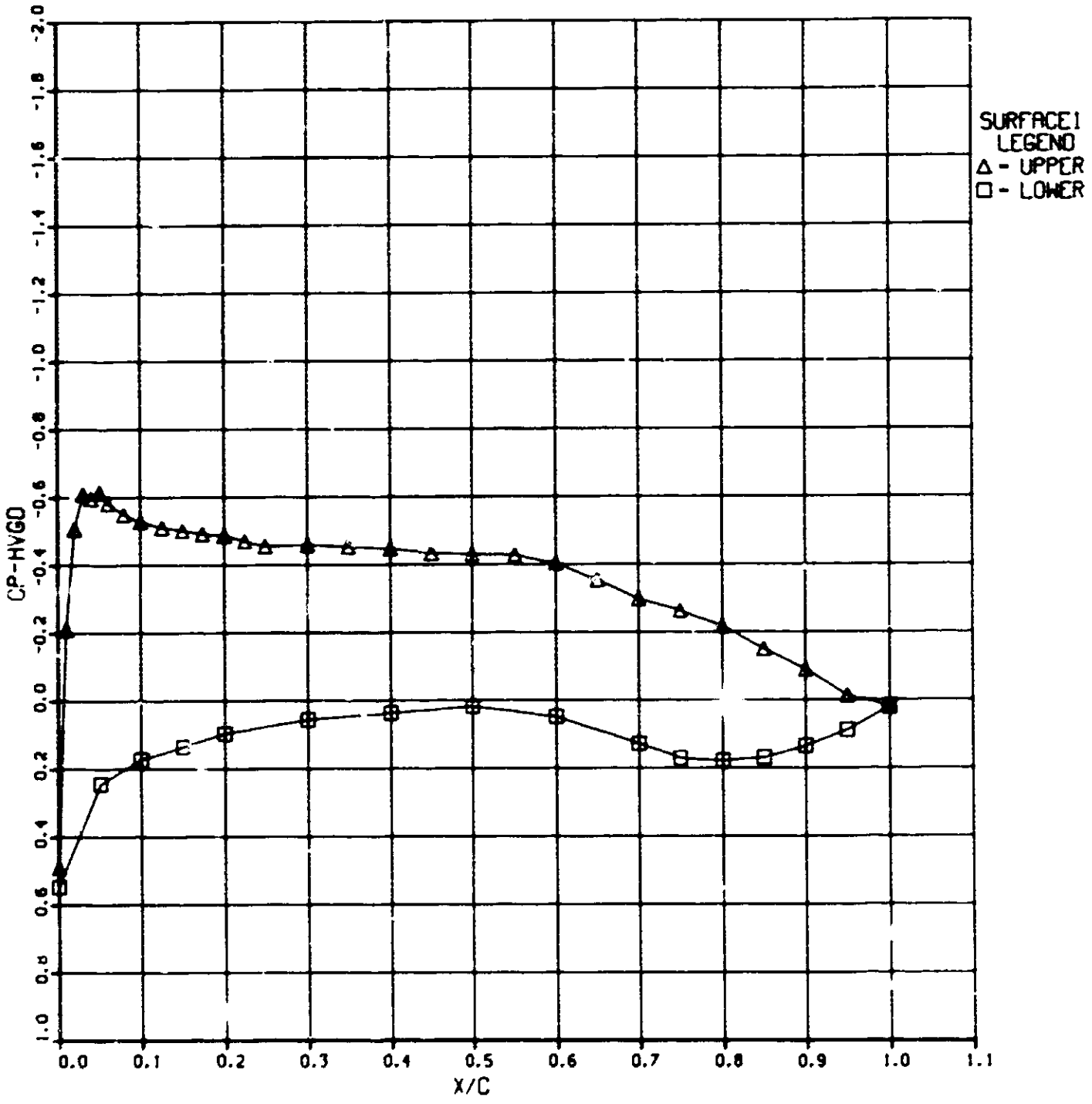


356-1-66 169.00: 2.00 CONF-17 MACH-0.850 RN-4.377 PT-2253 ALPHA- 5.00



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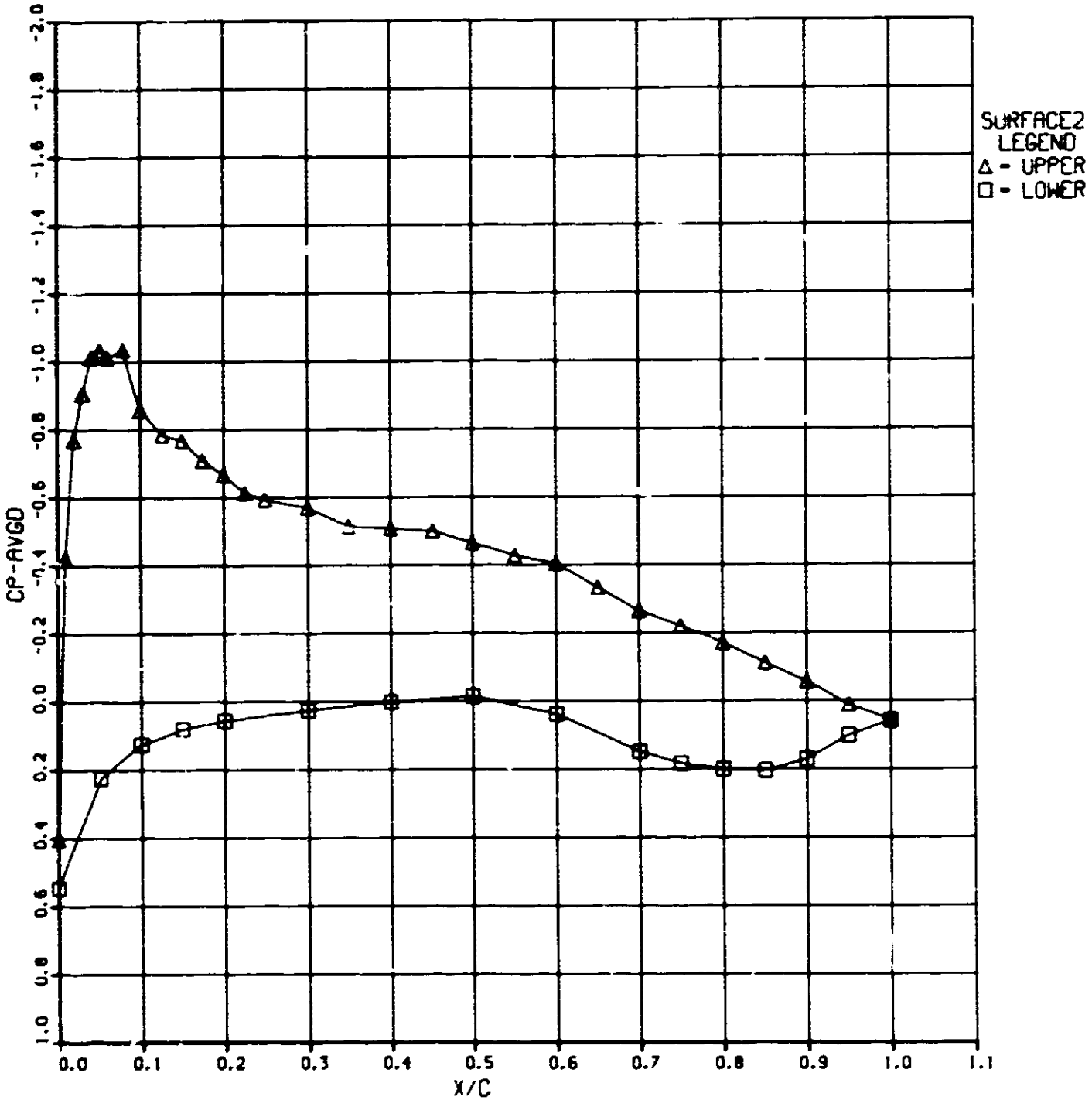

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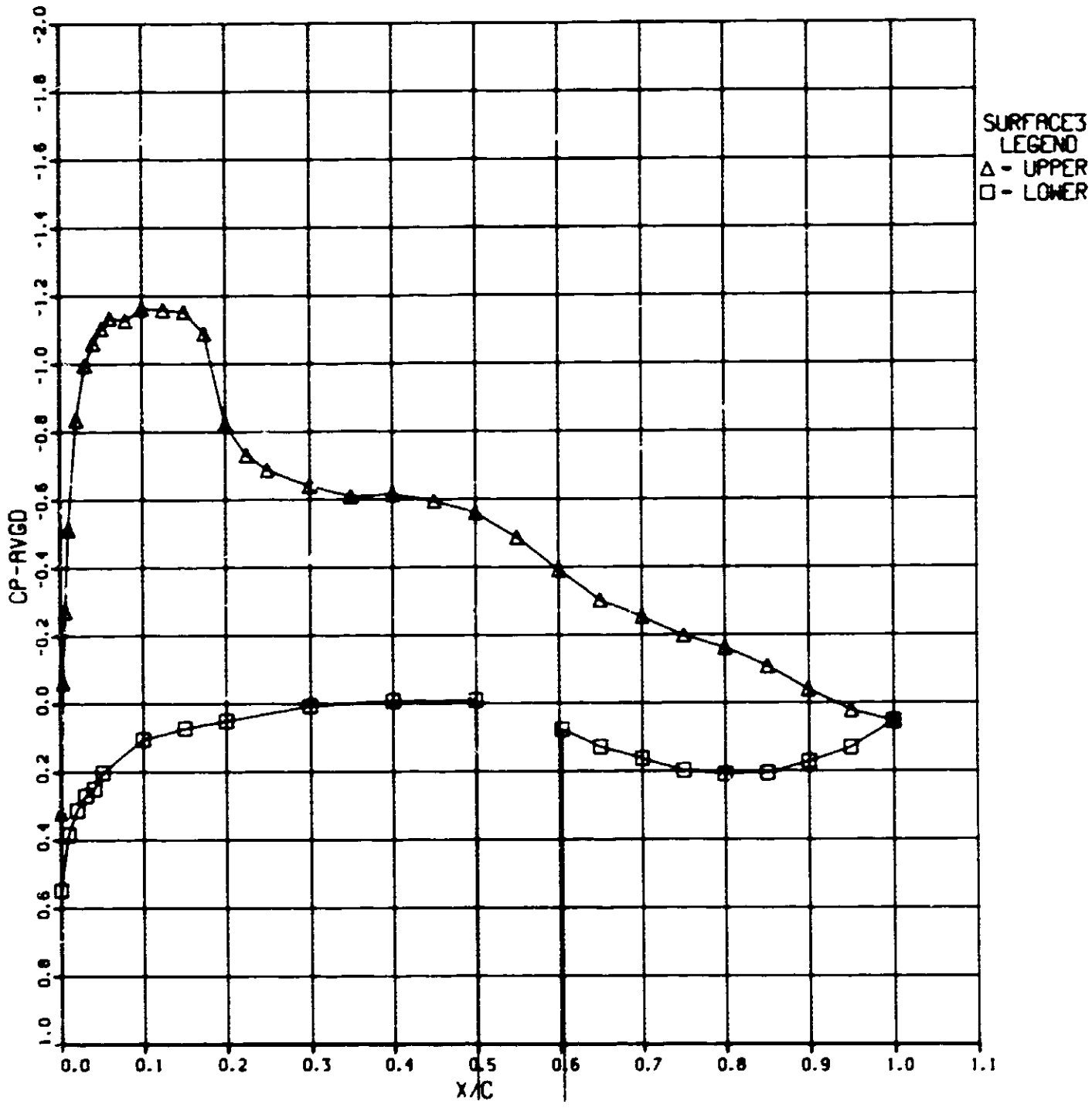

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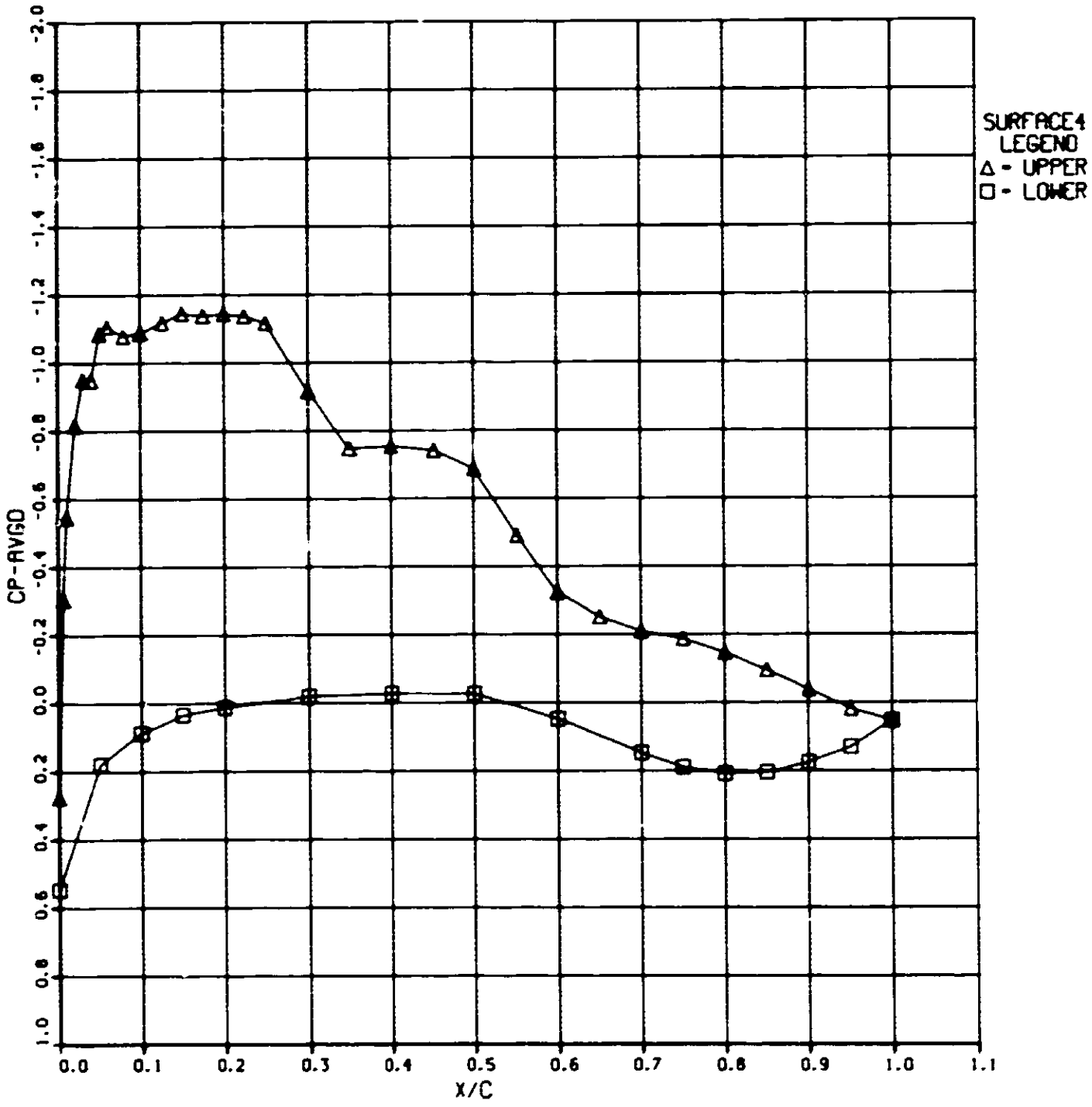

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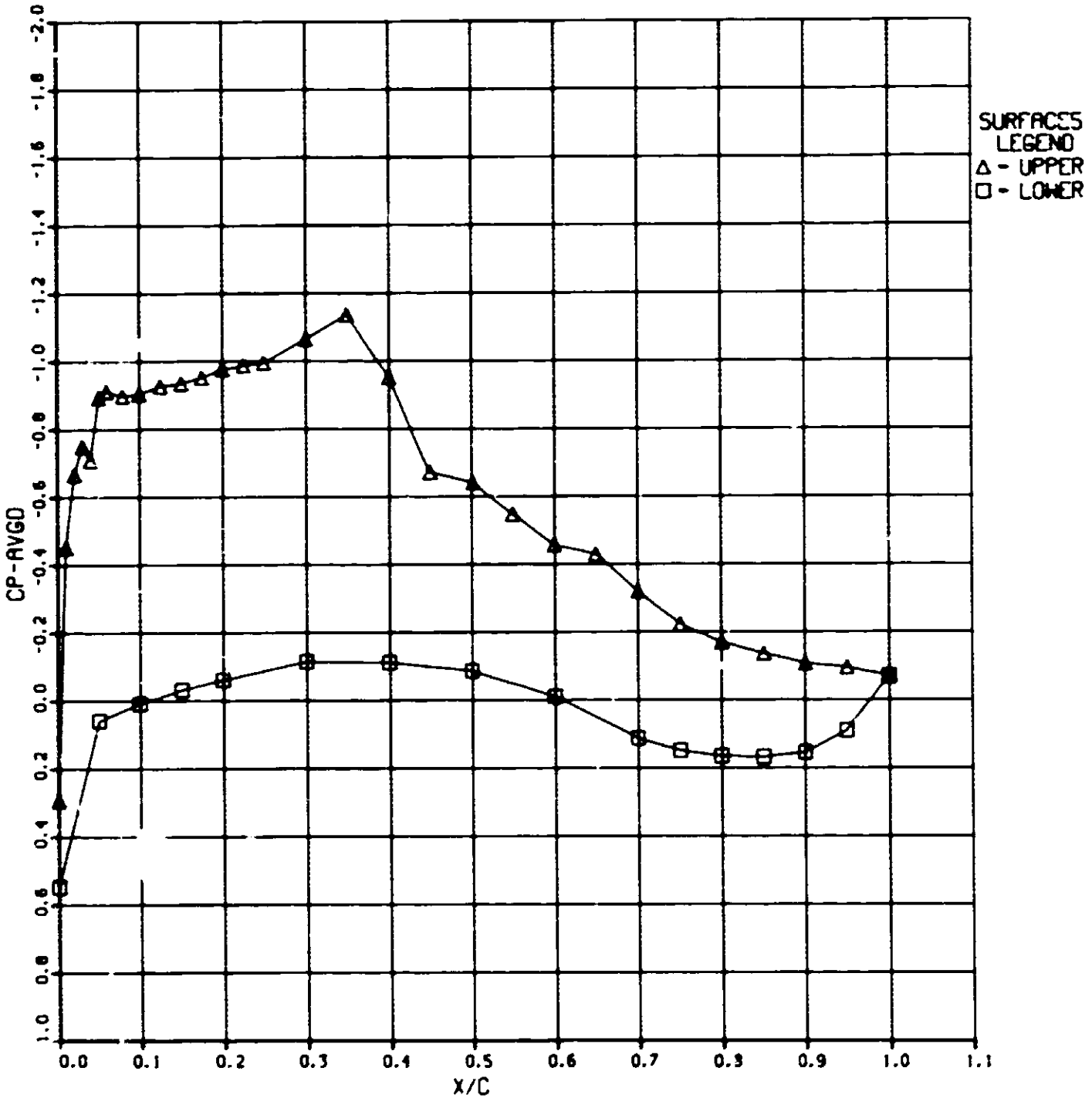

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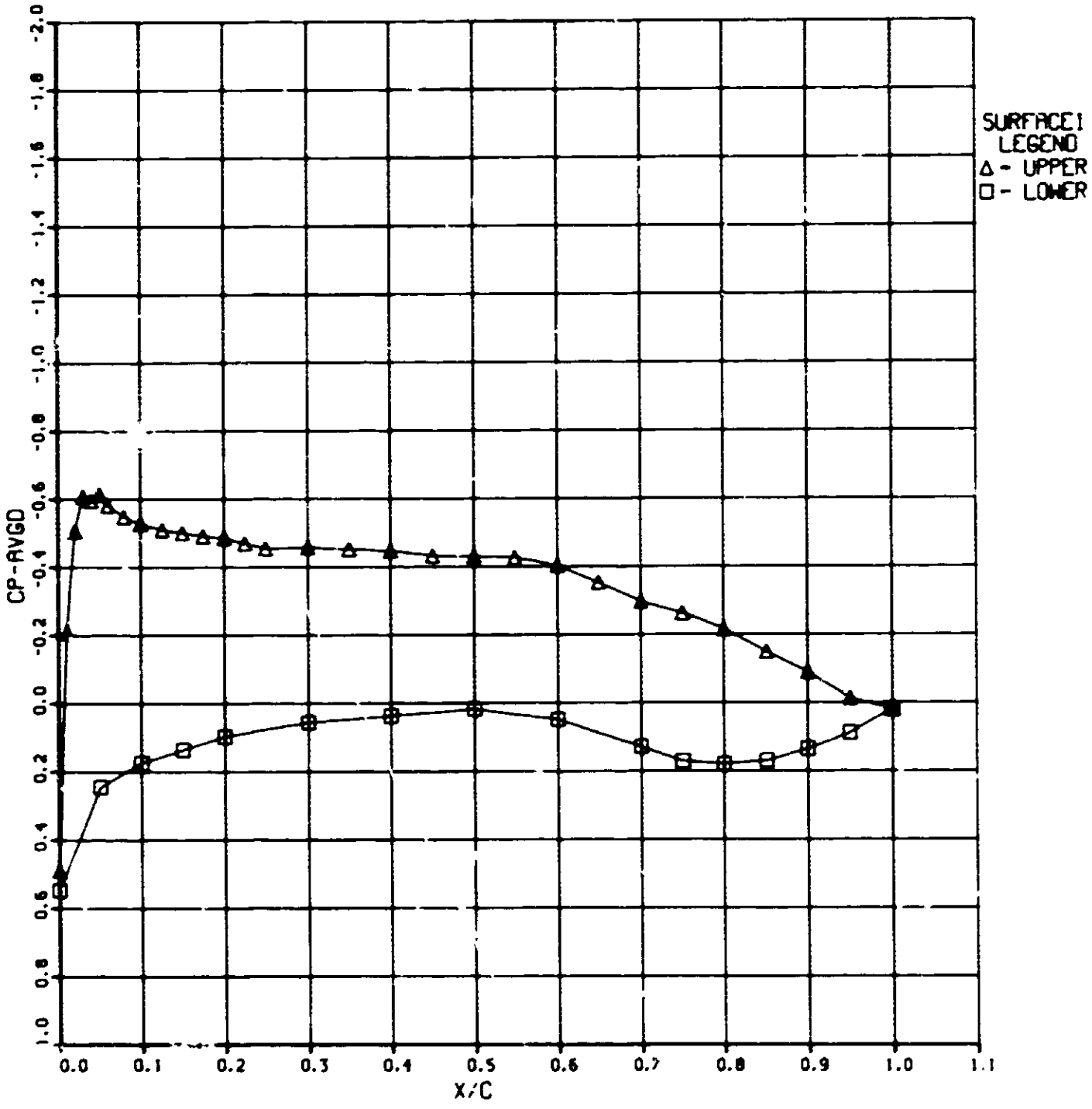

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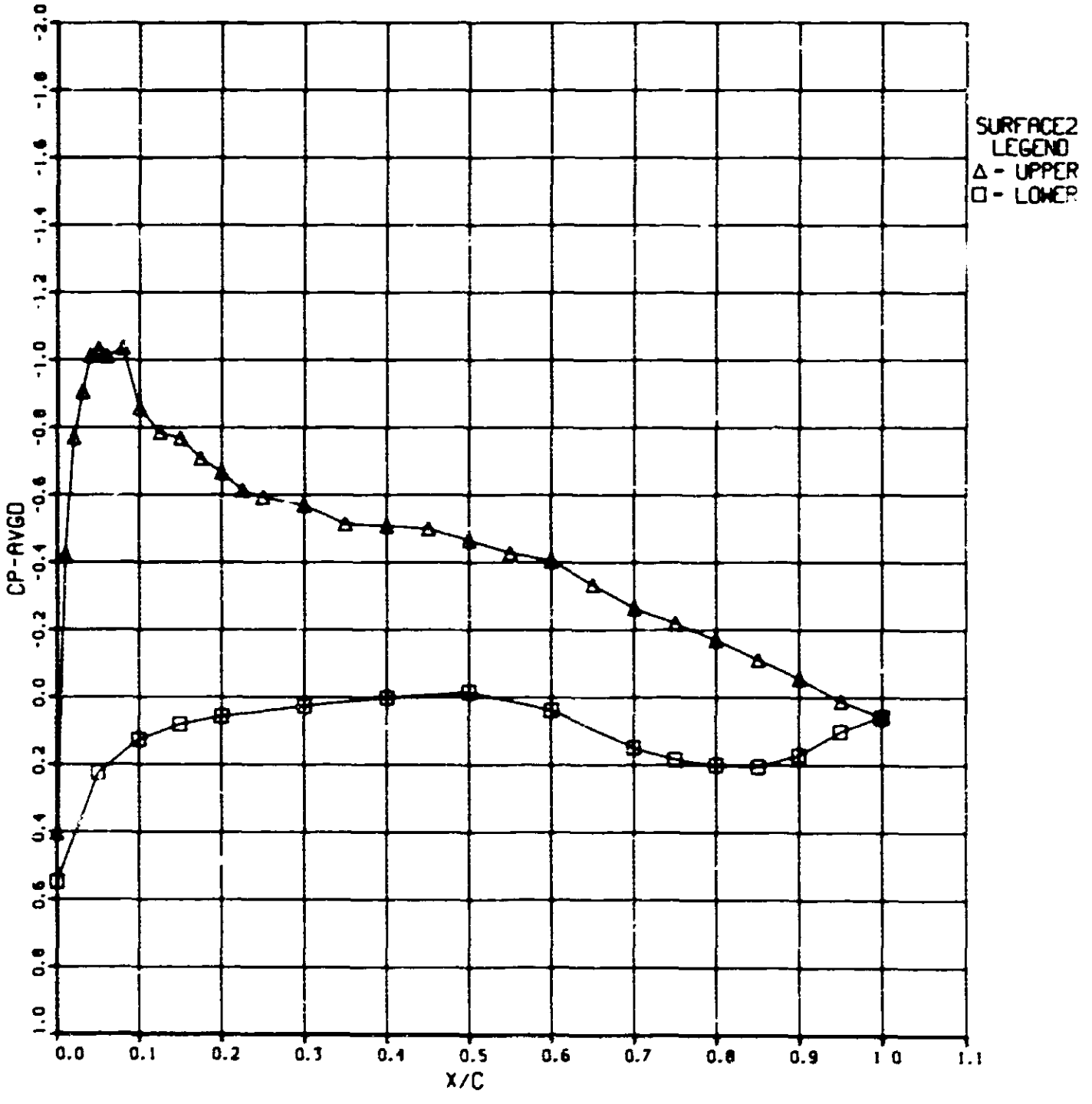

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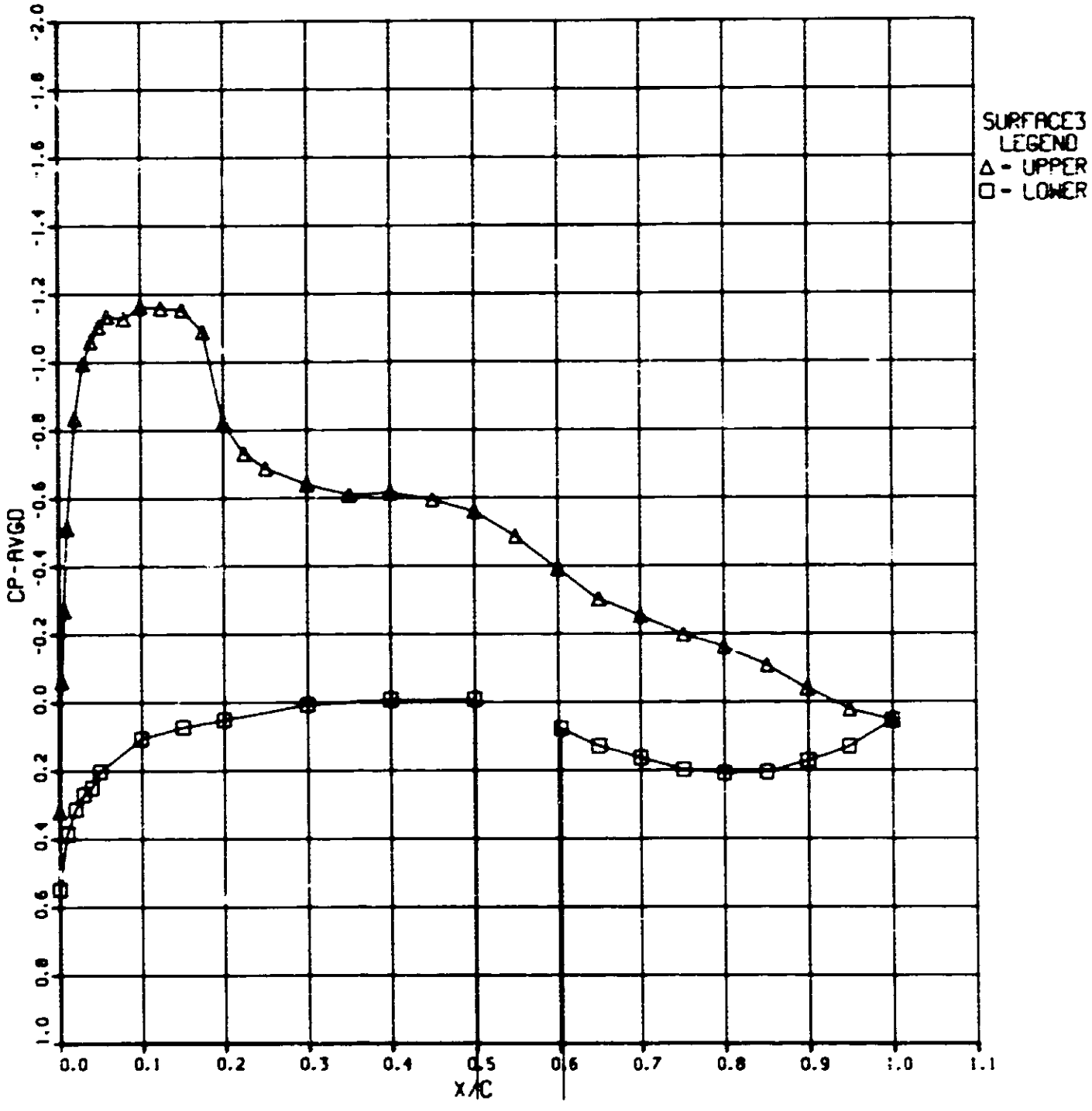

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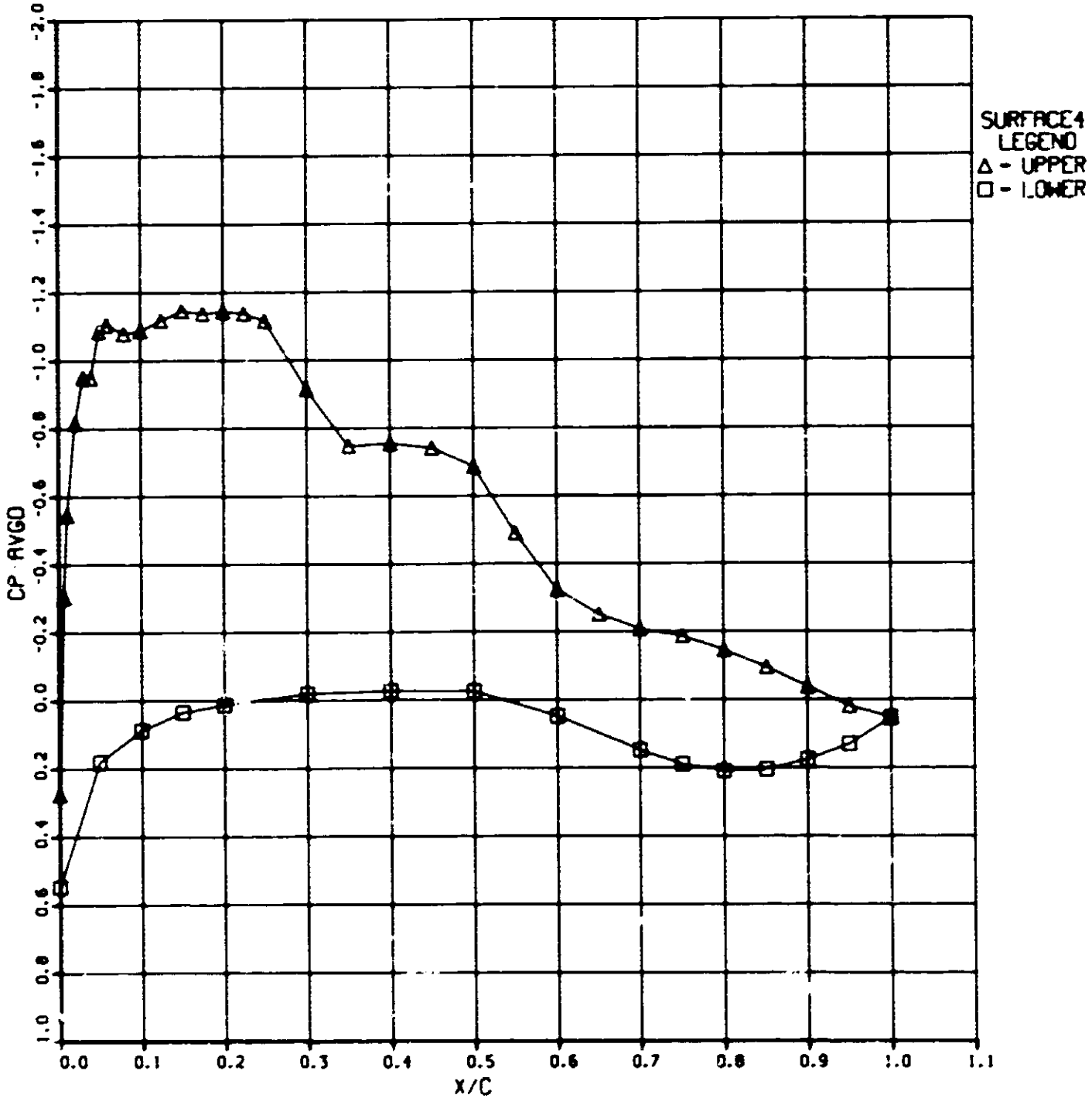
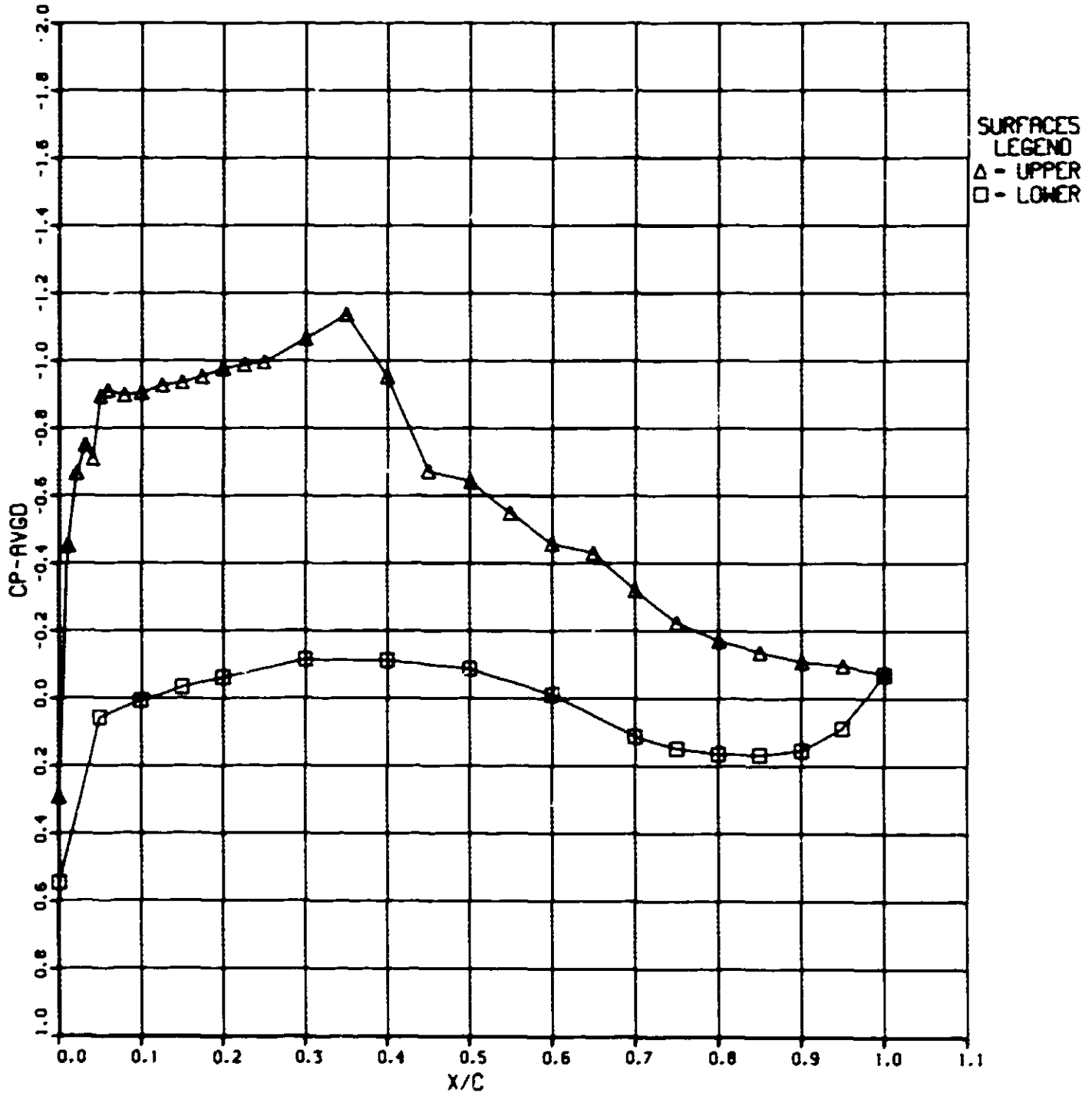


356-1-66 169.00: 2.00 CONF-17 MACH-0.850 RN-4.377 PT-2253 ALPHA- 5.00



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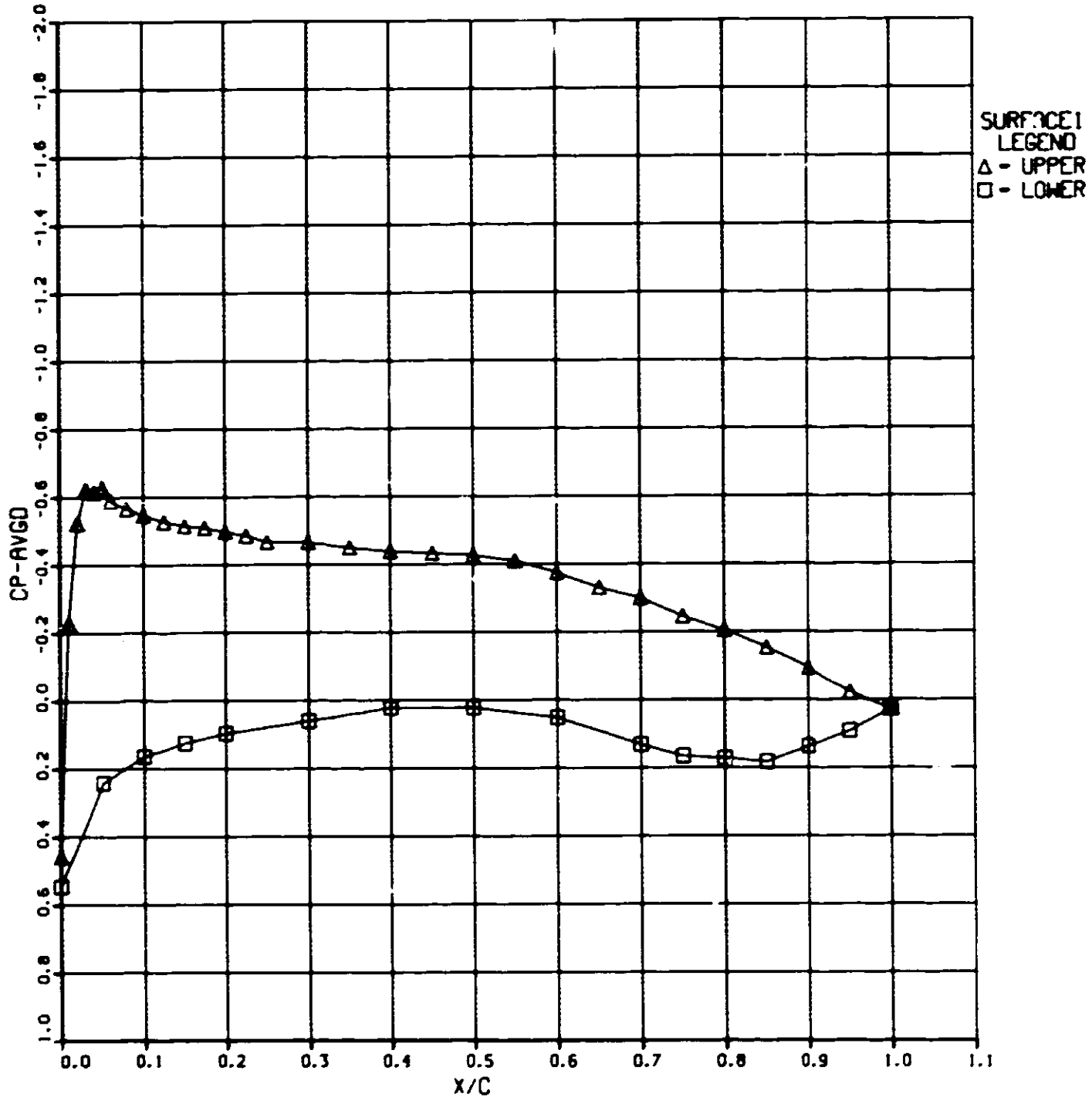

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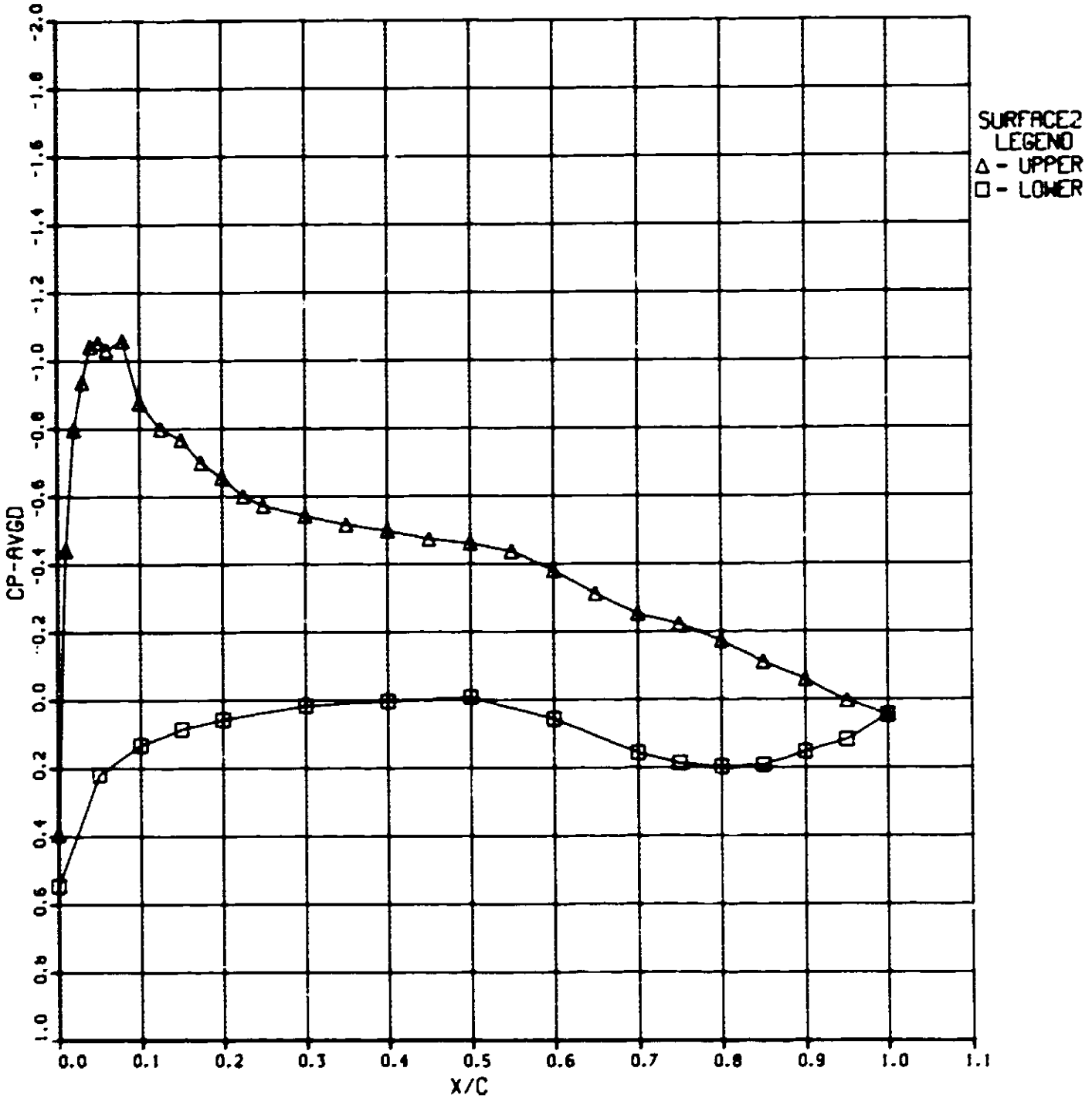

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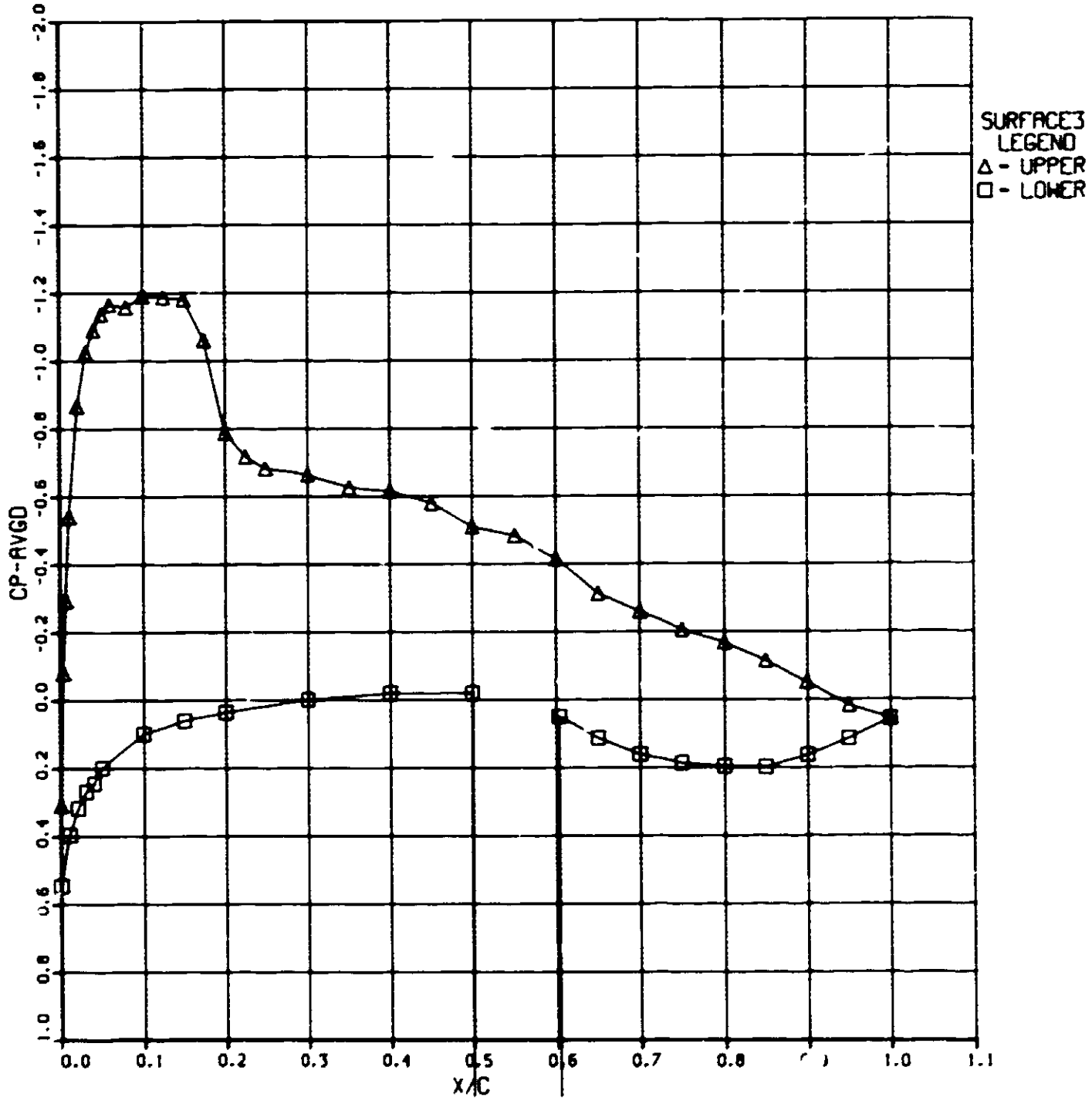

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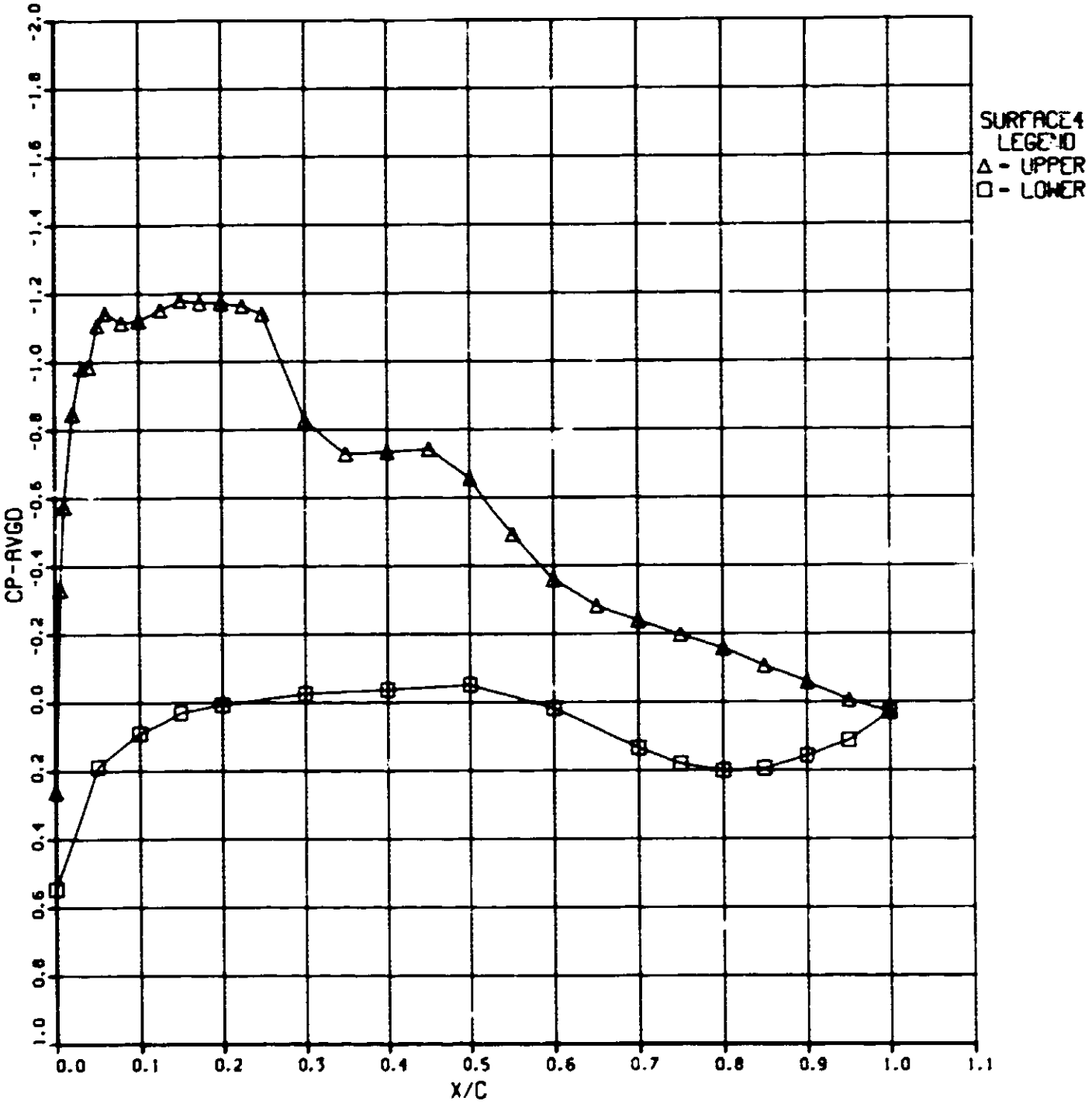

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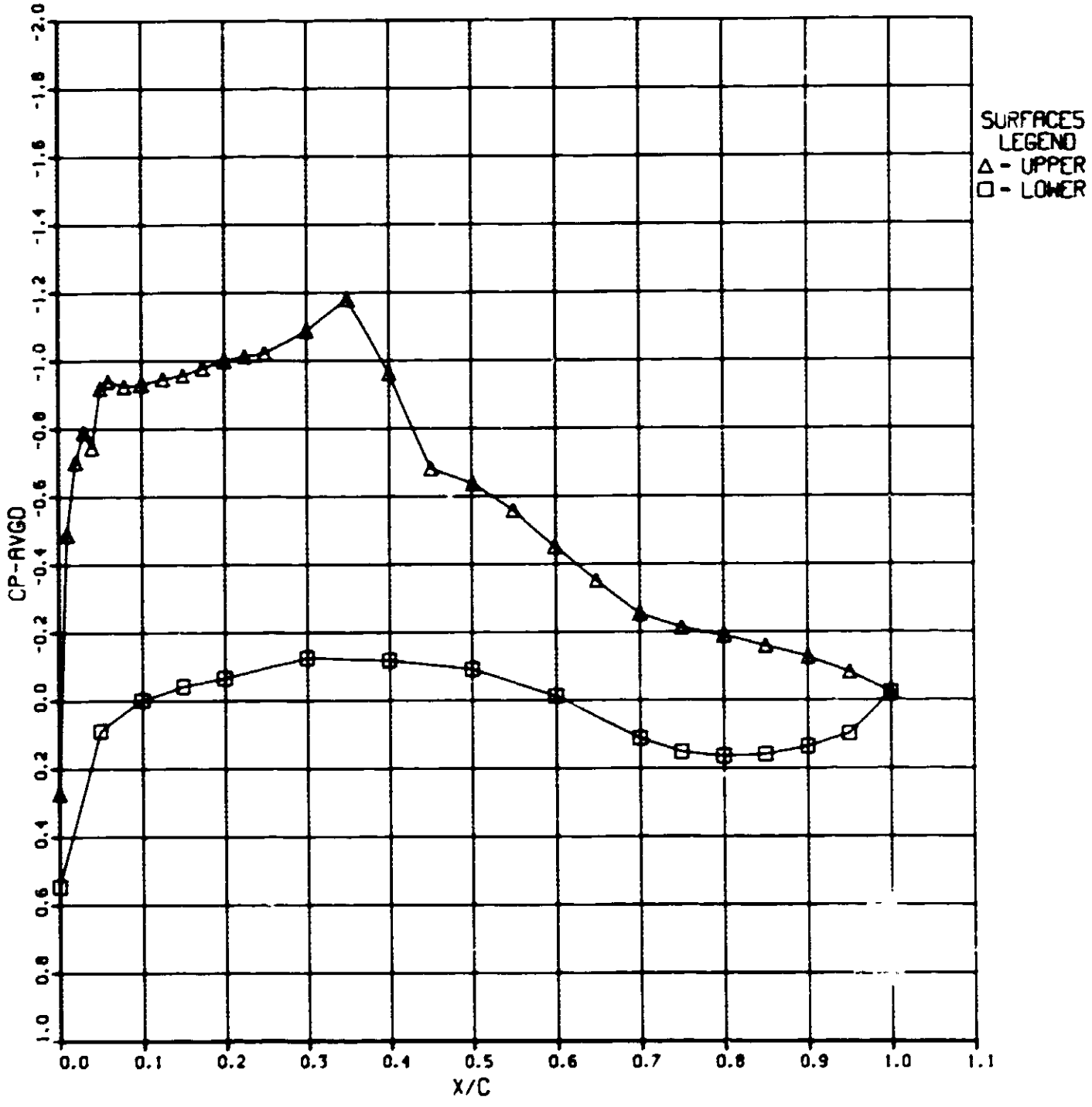

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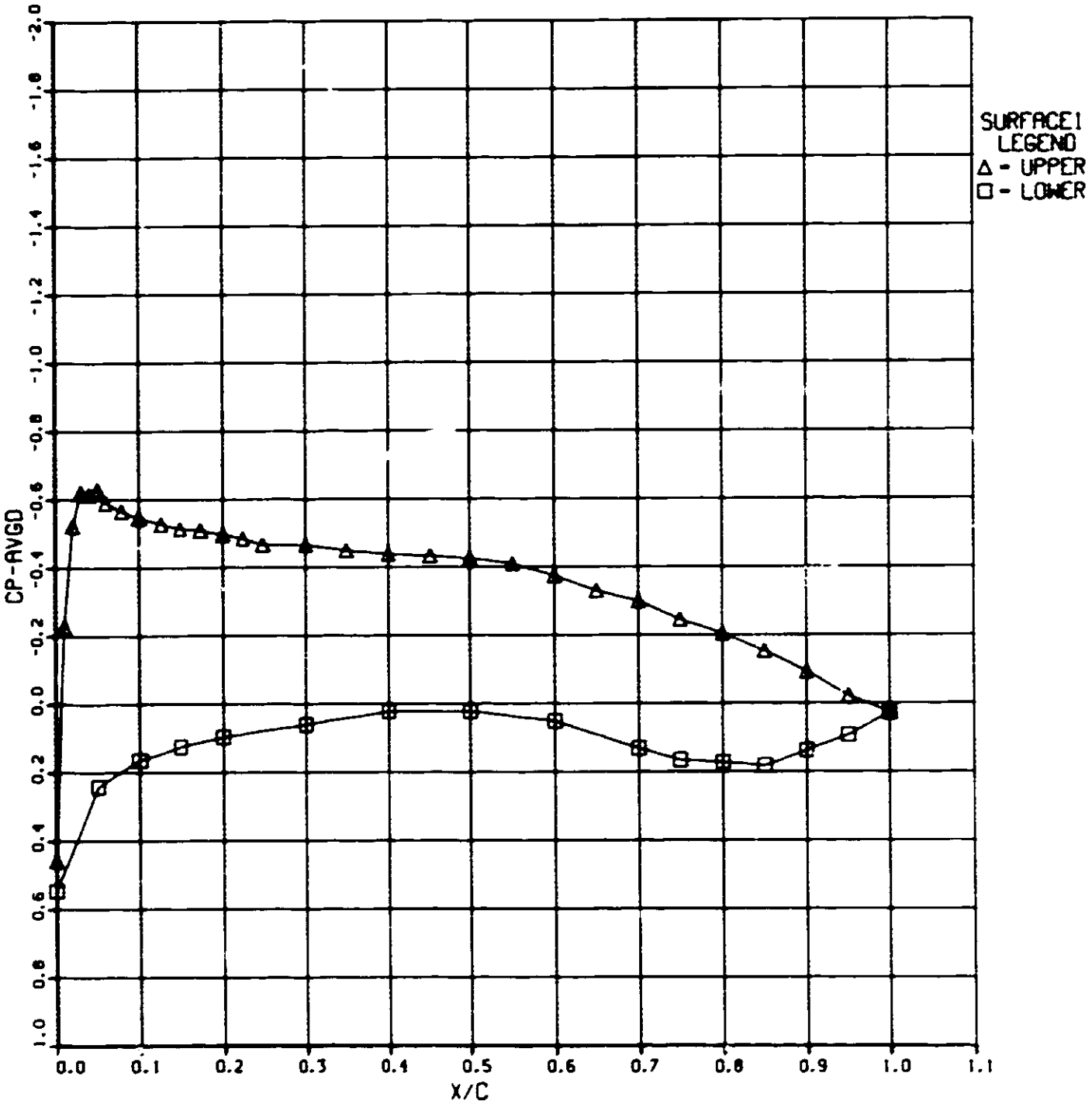

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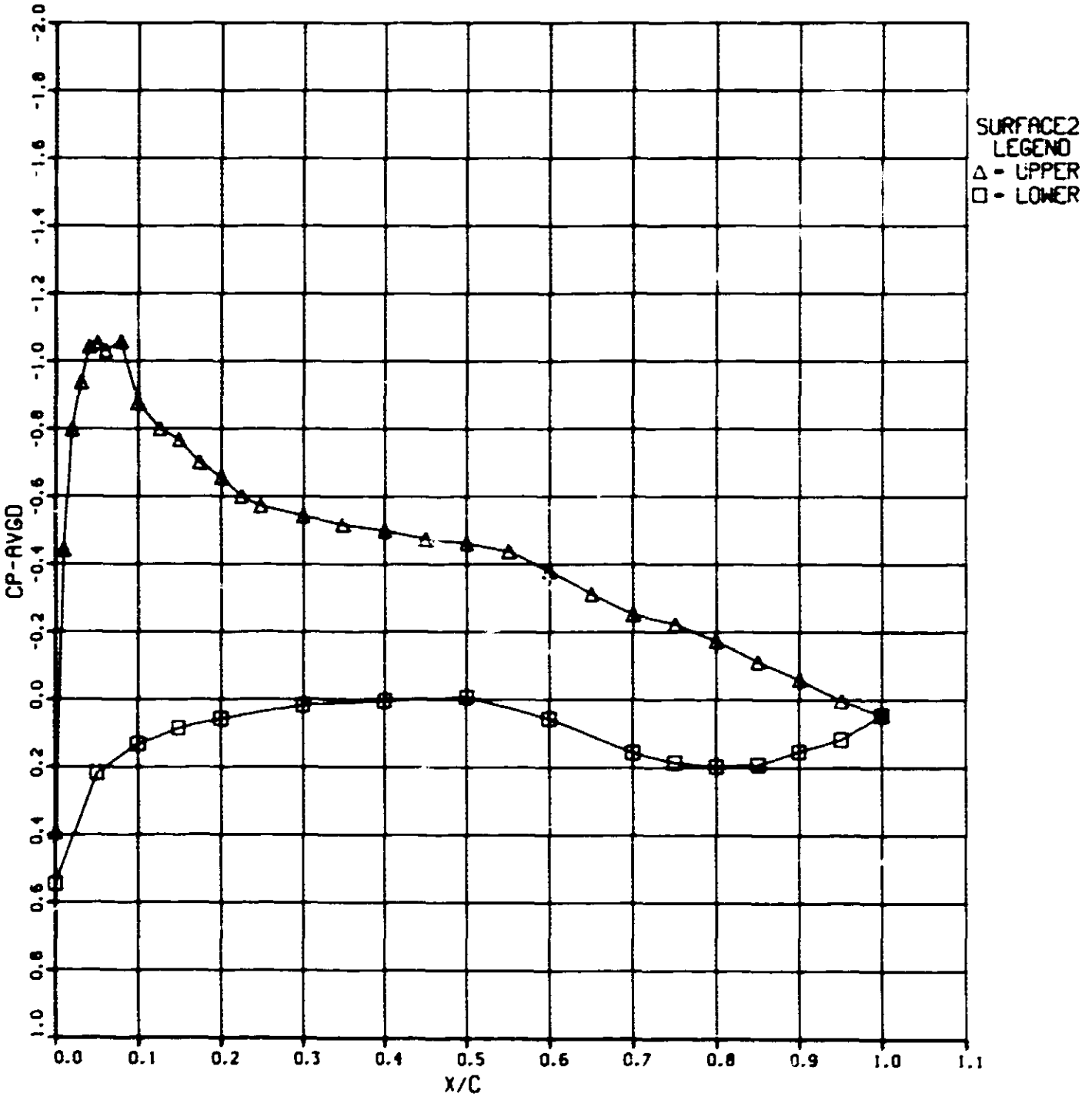

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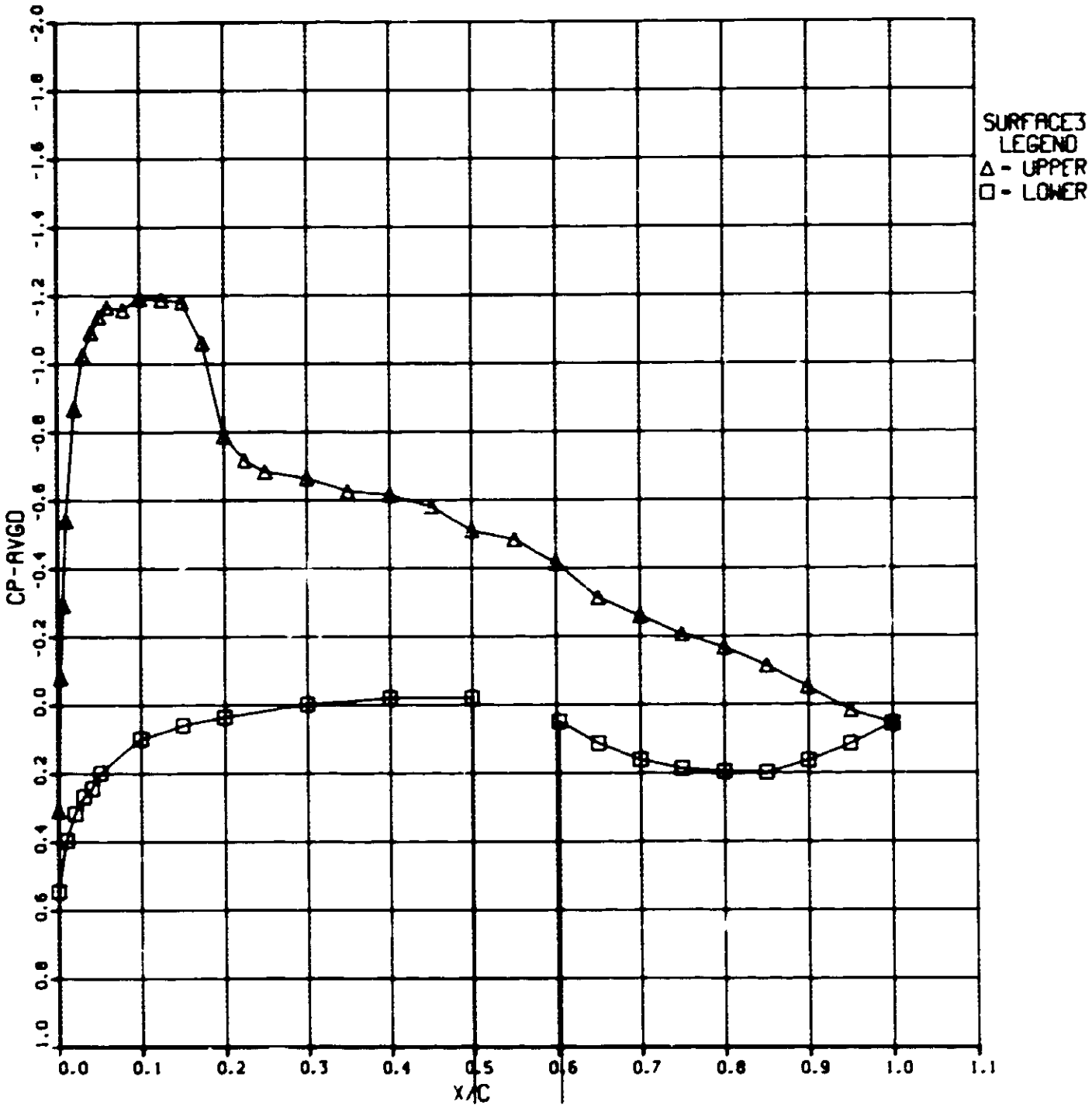

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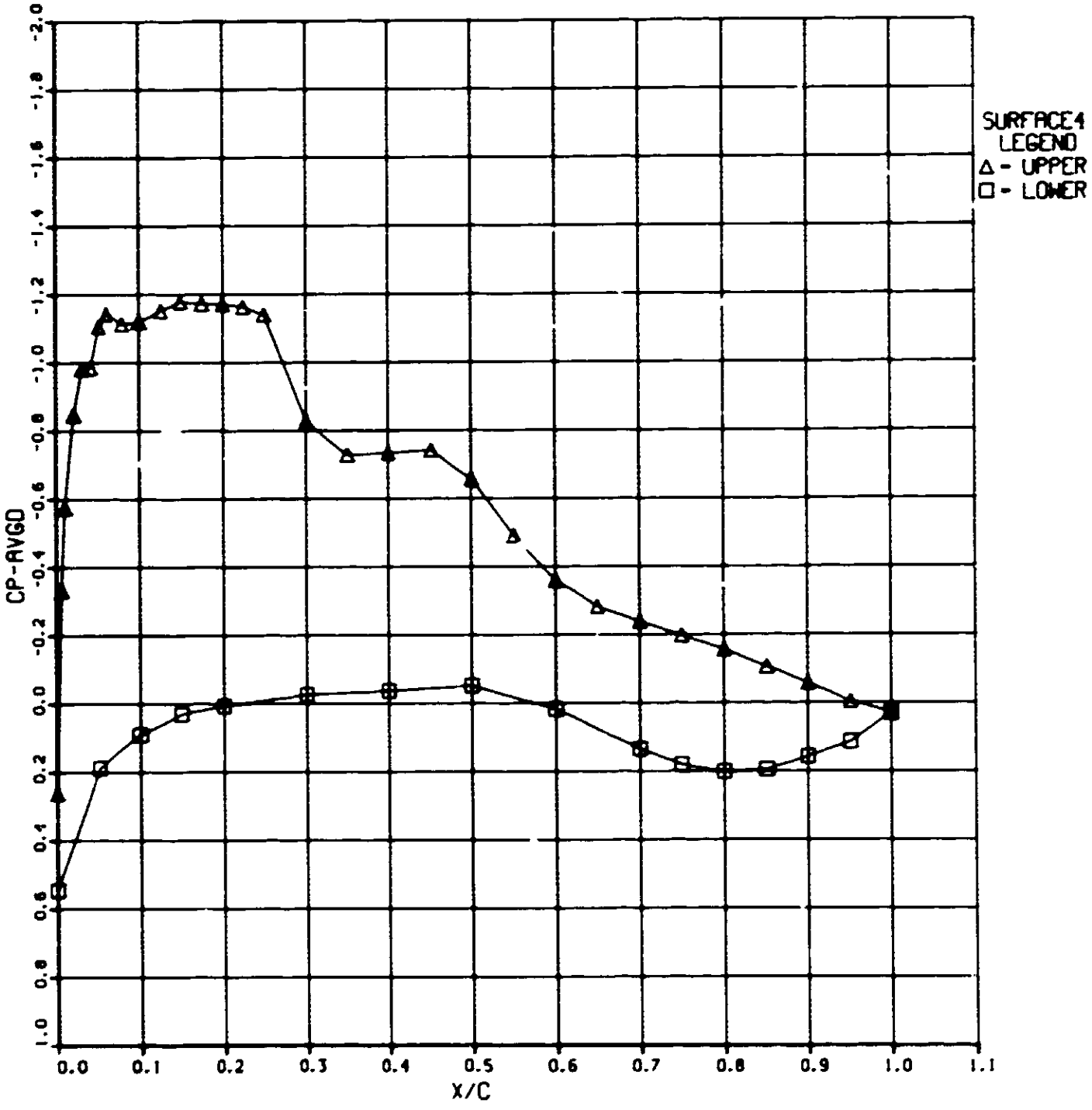
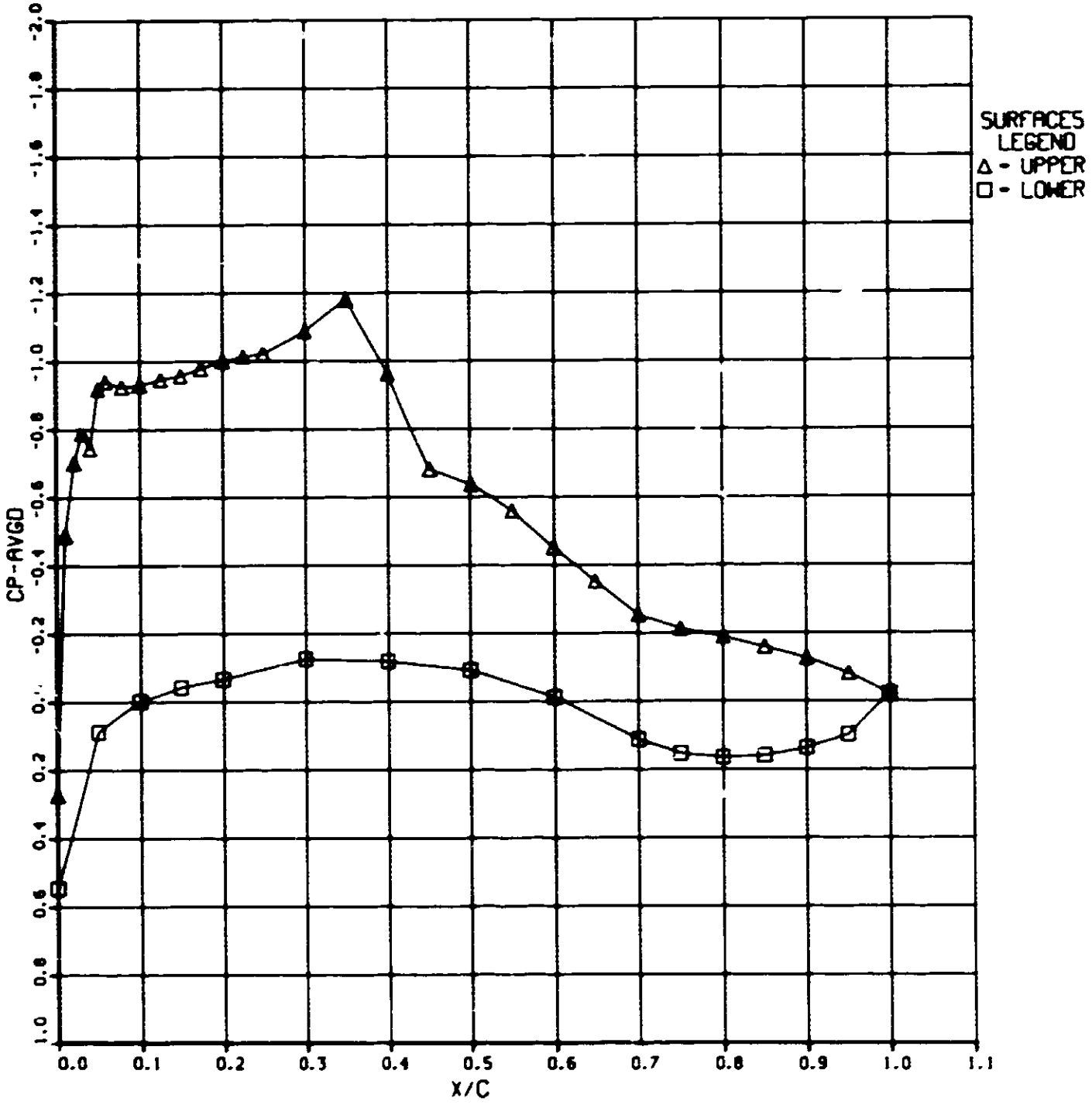
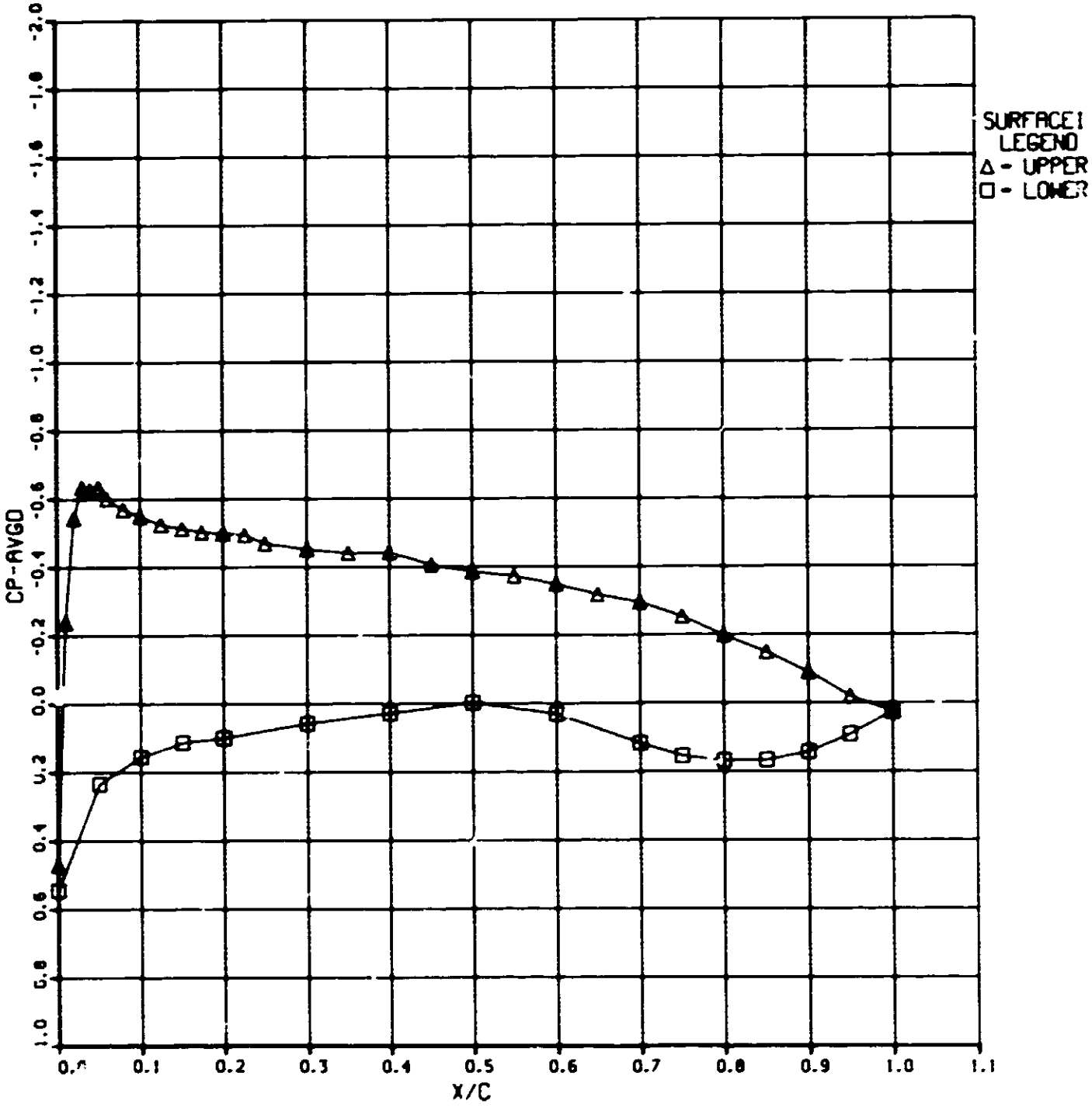


356-1-S6 170.00: 2.00 CONF-17 MACH-0.837 RN-4.382 PT-2276 ALPHA- 5.00



356-1-66 171.00: 2.00 CONF-17 MACH-0.829 RN-4.382 PT-2289 ALPHA- 5.00



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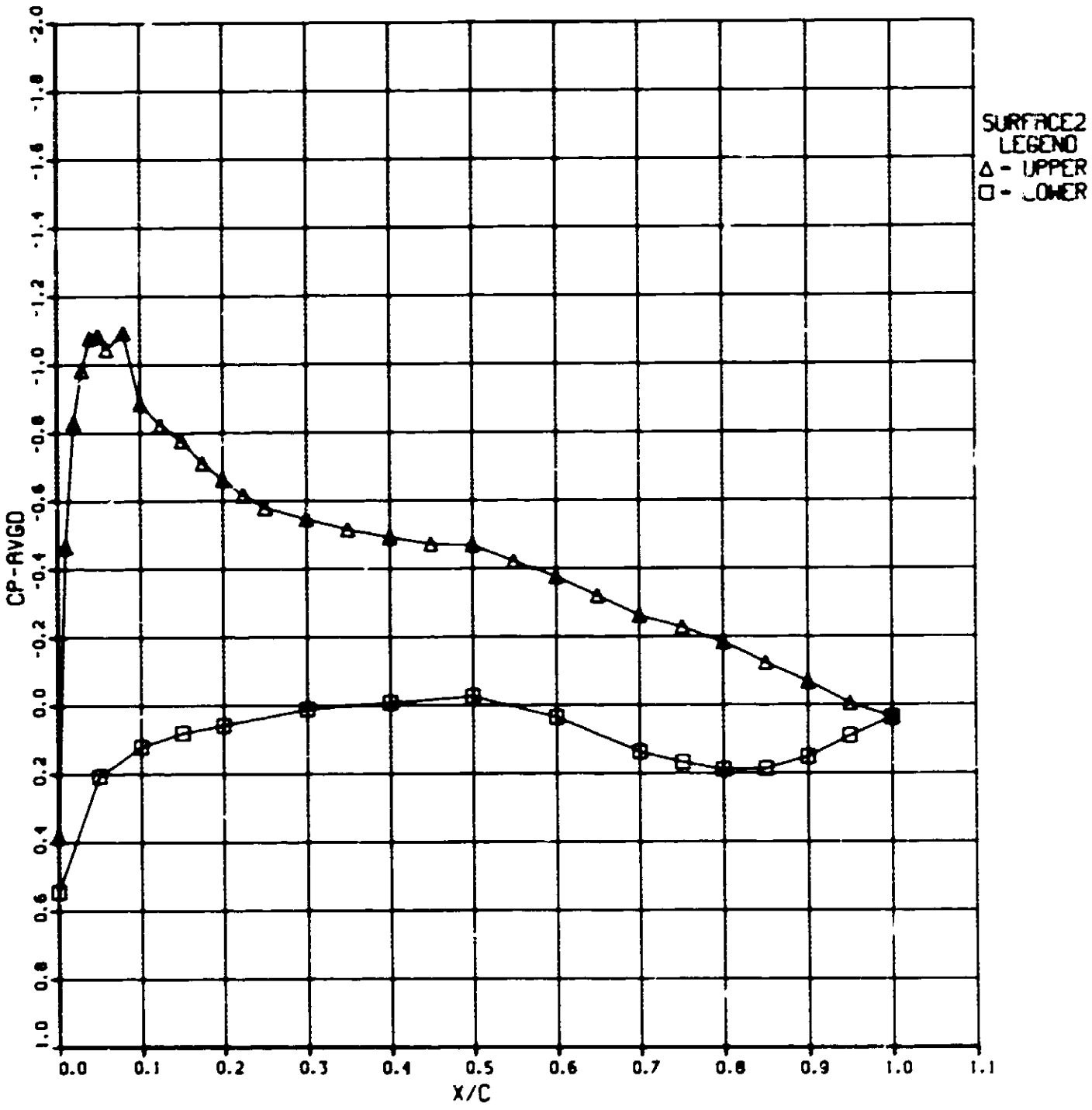

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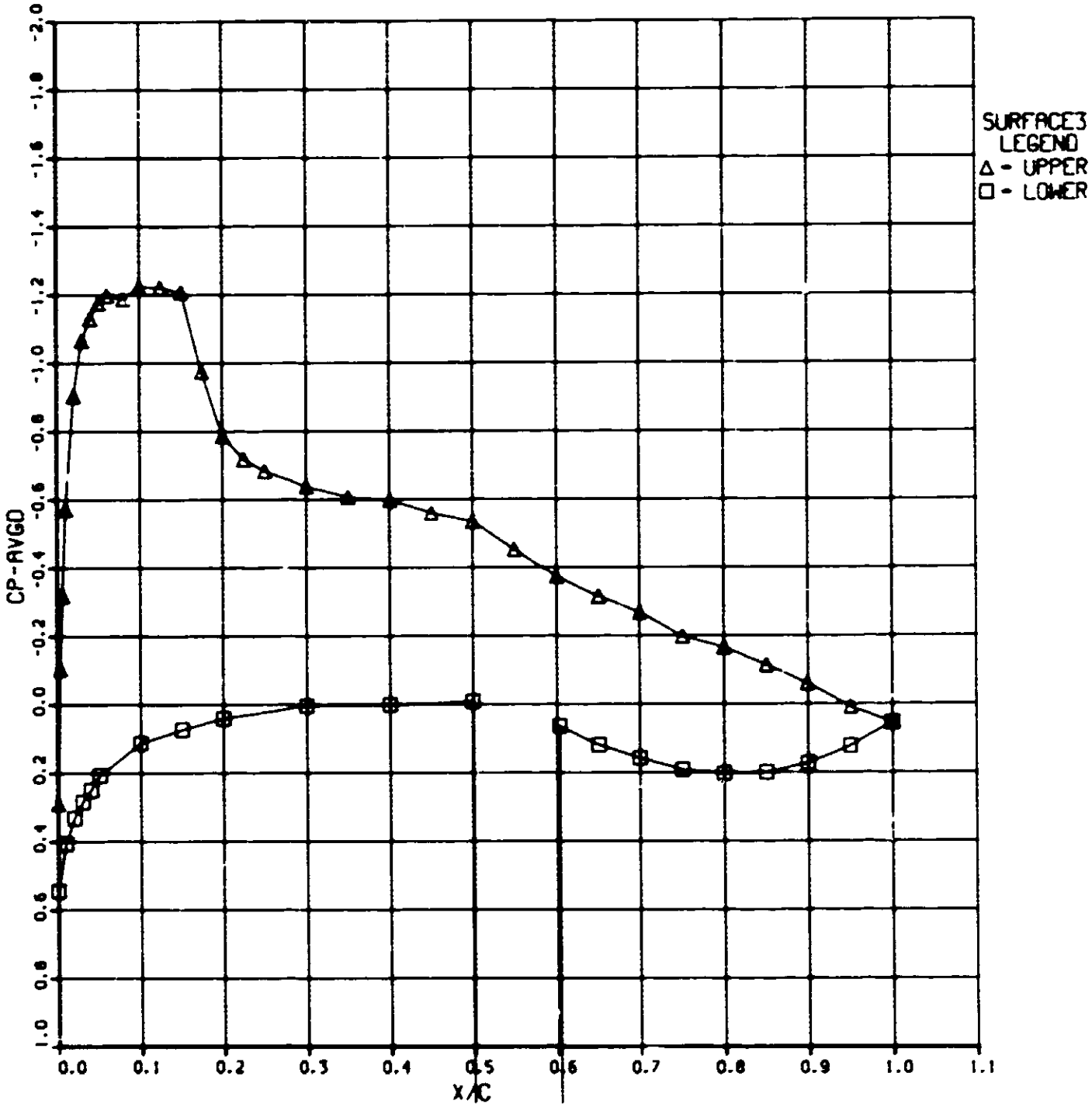

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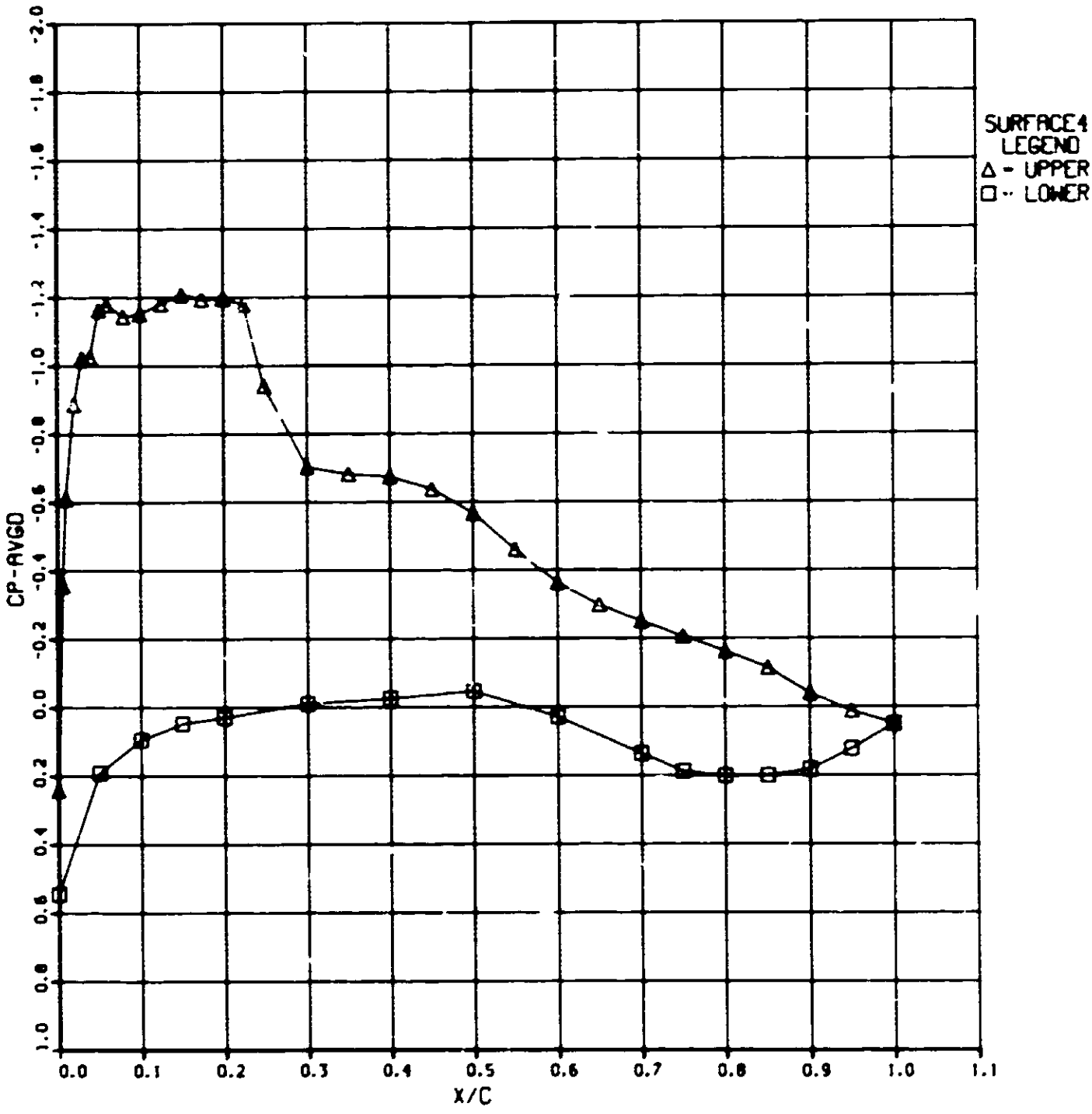

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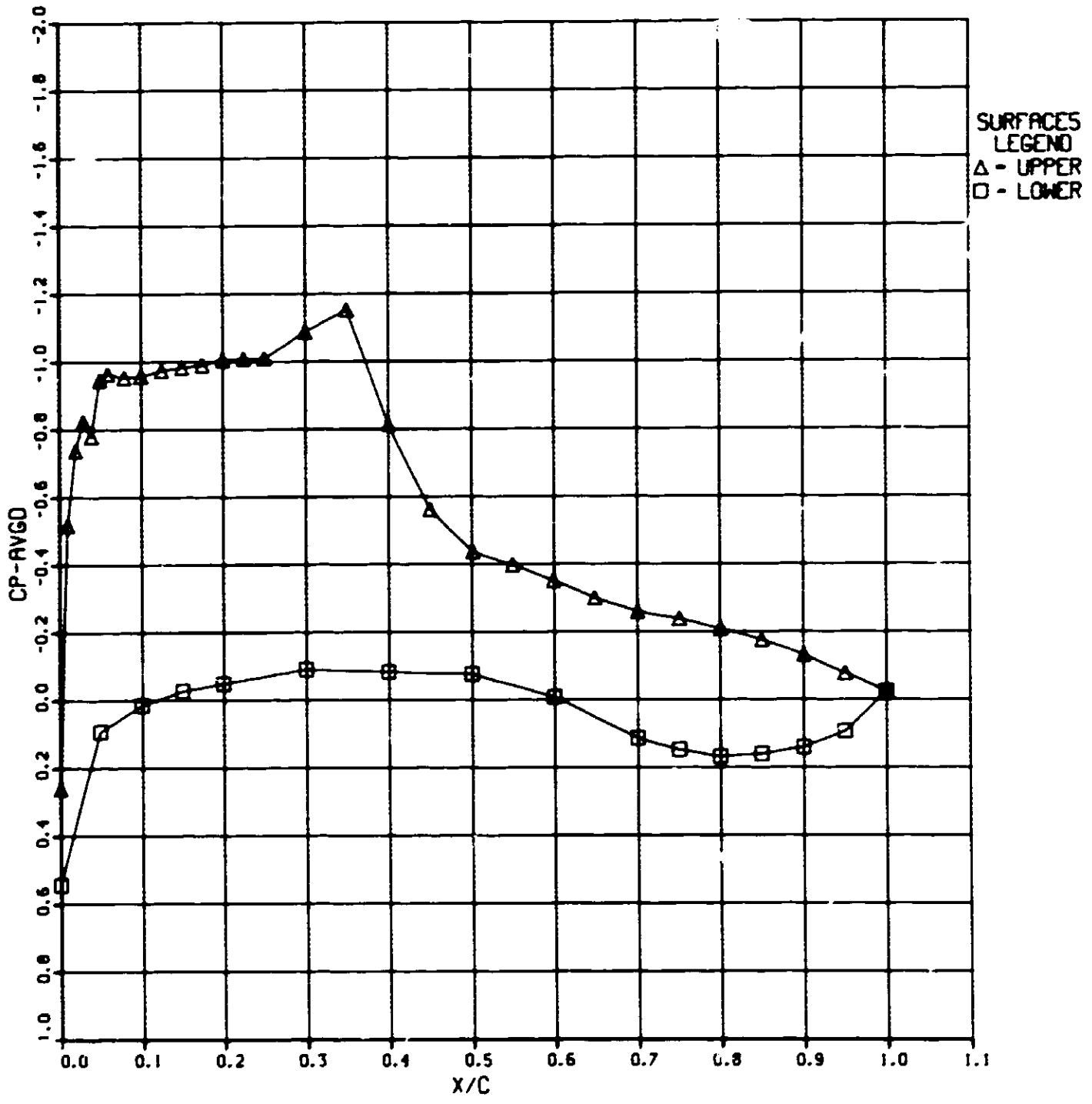

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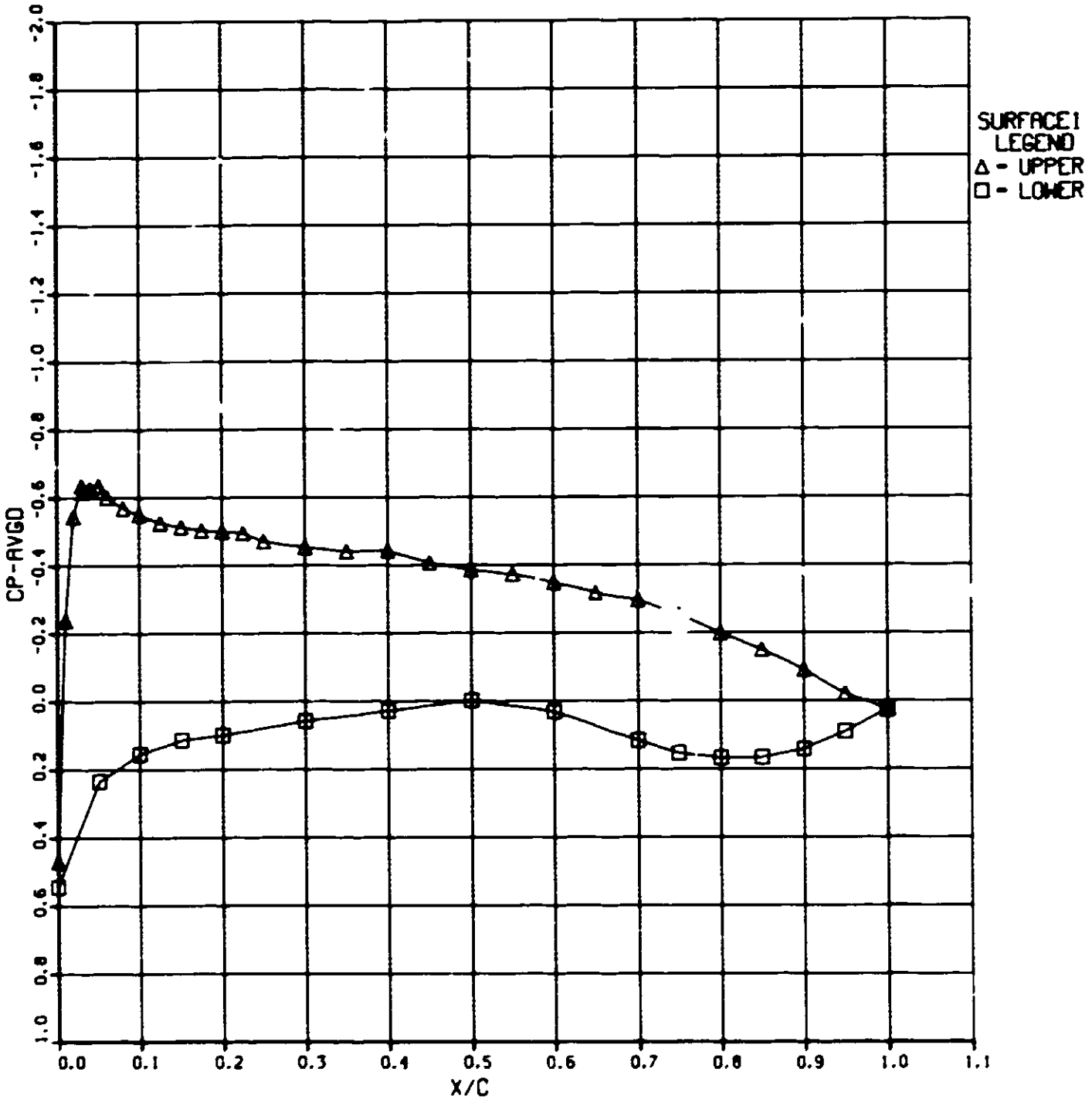

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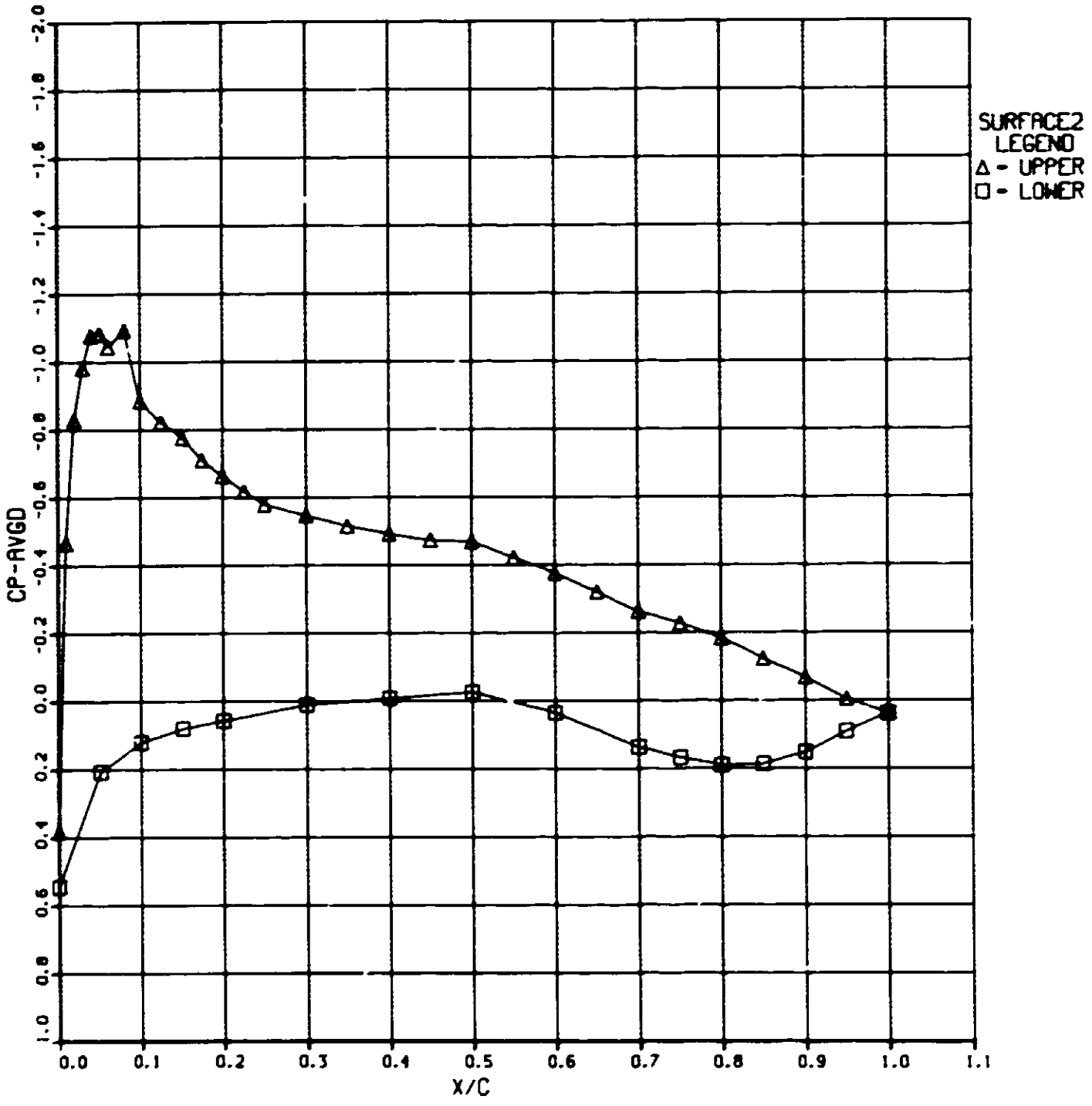

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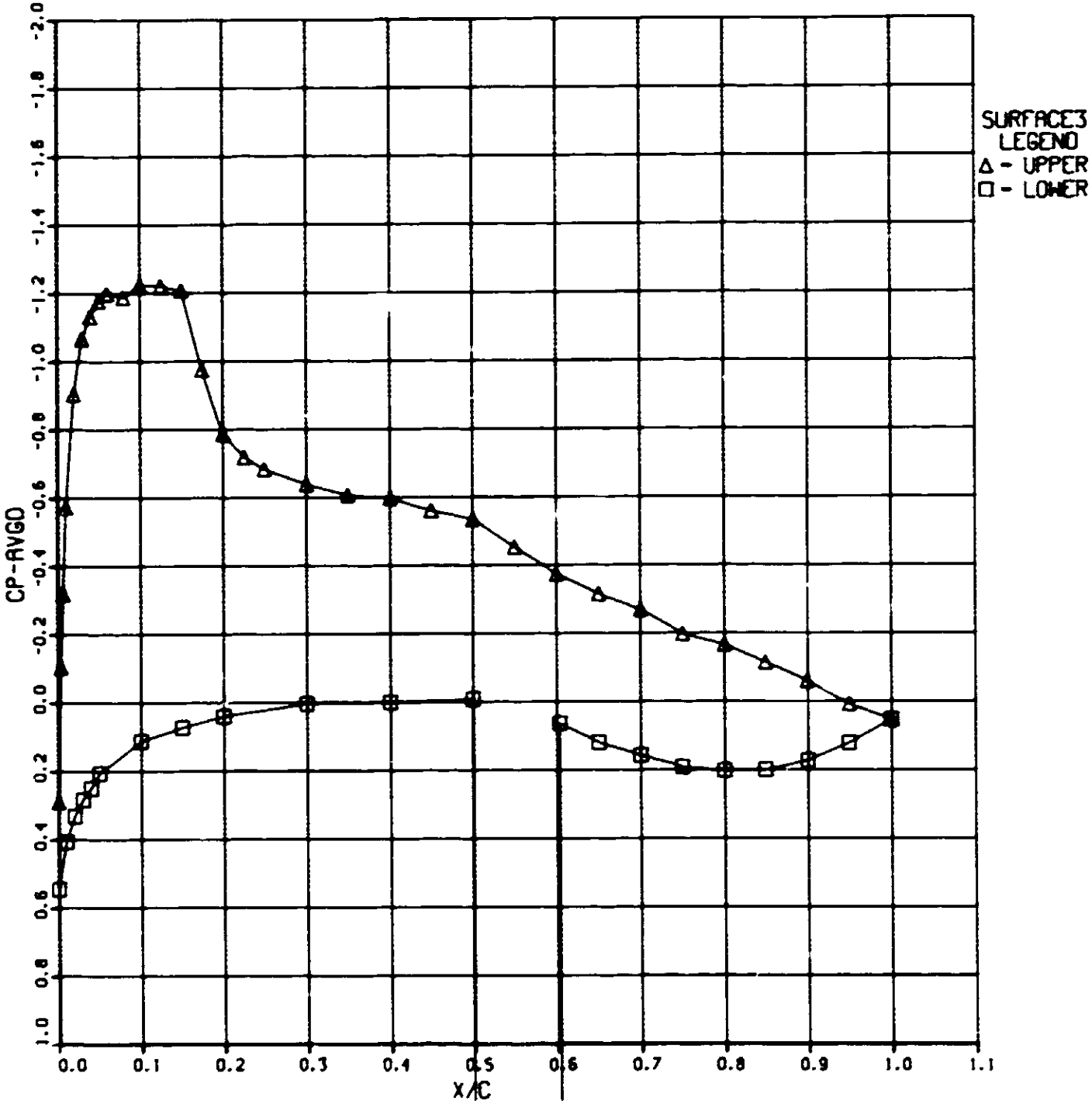

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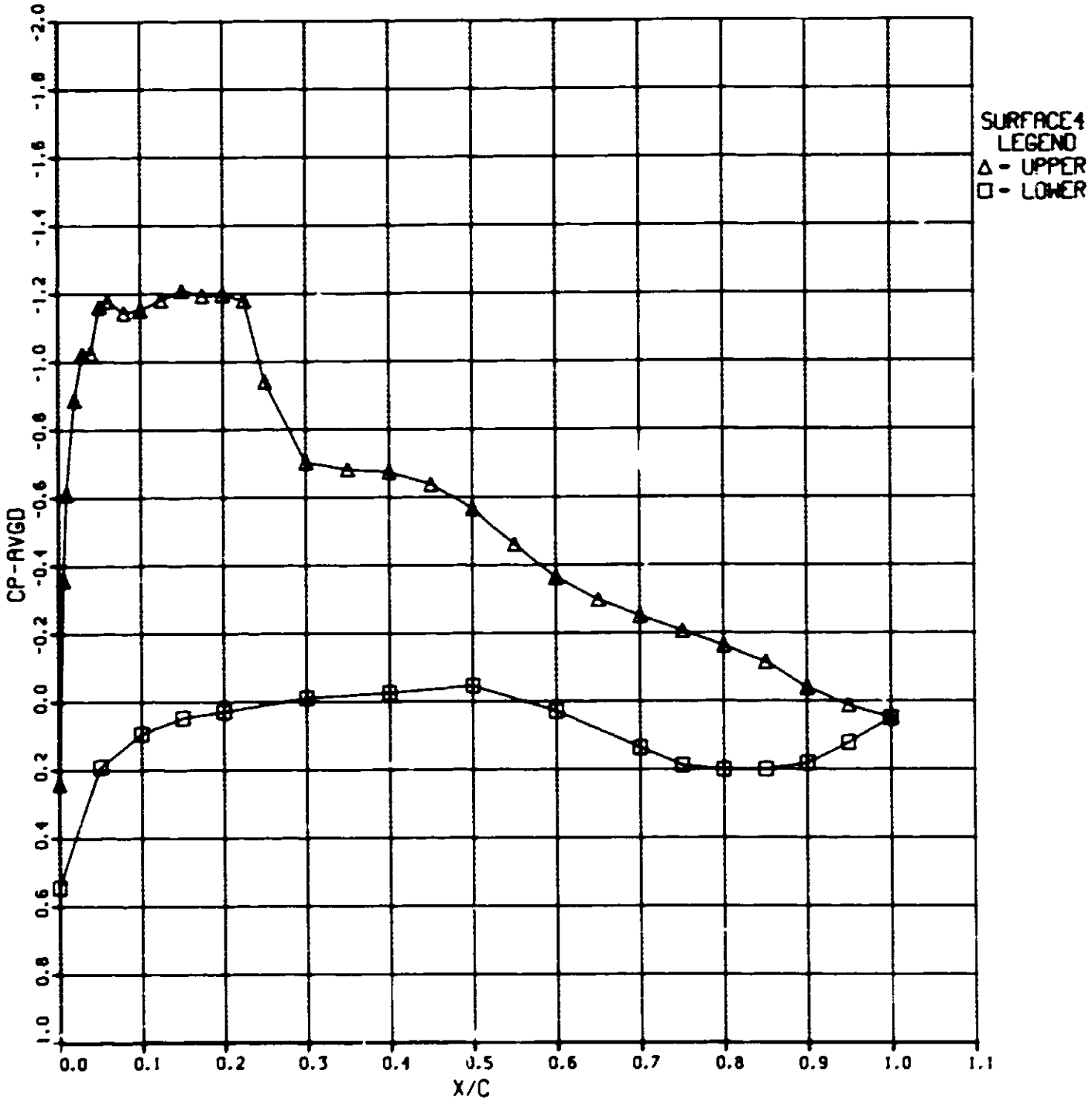
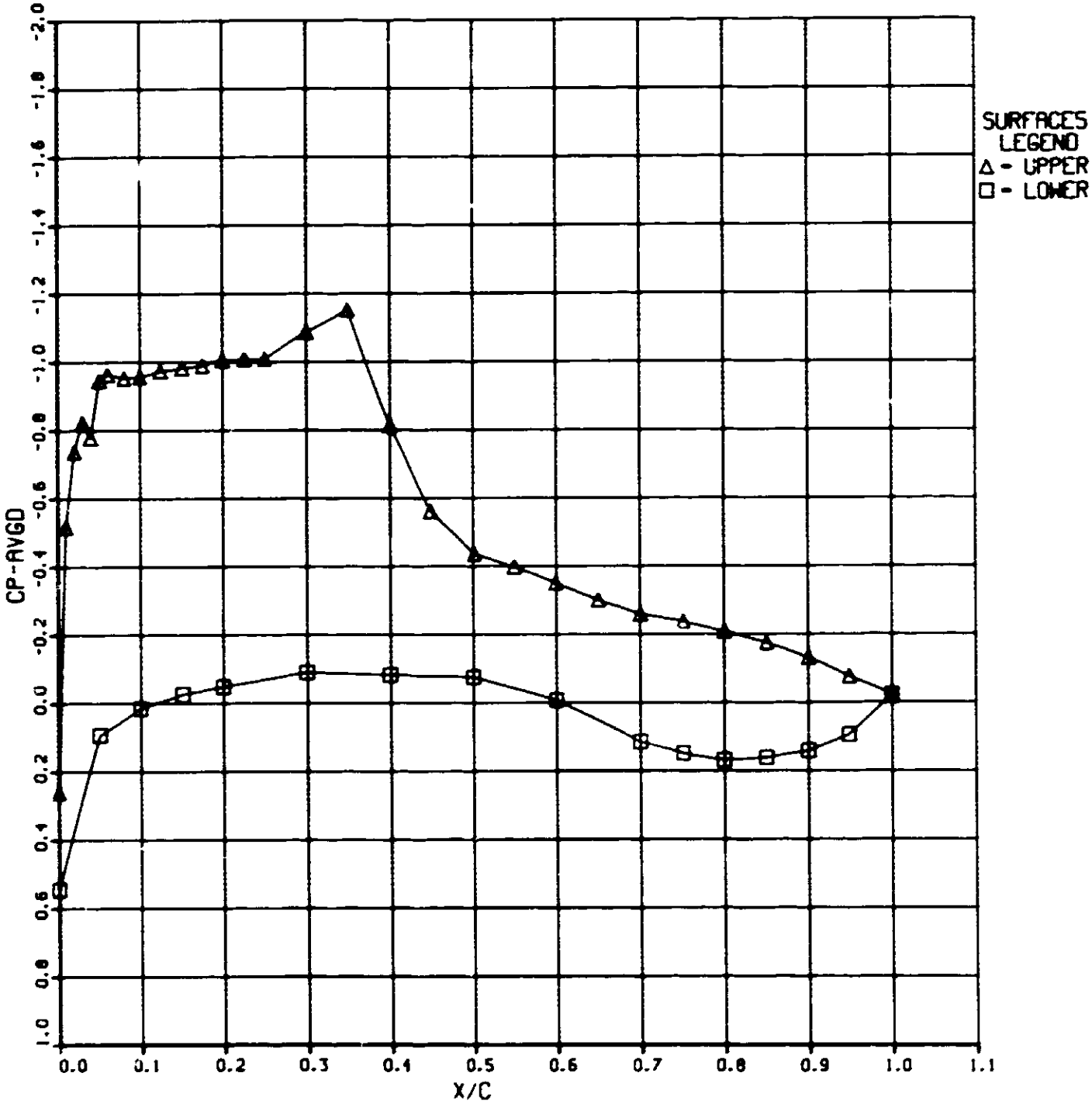


356-1-66 171.00: 2.00 CONF-17 MACH-0.829 RN-4.382 PT-2209 ALPHA- 5.00



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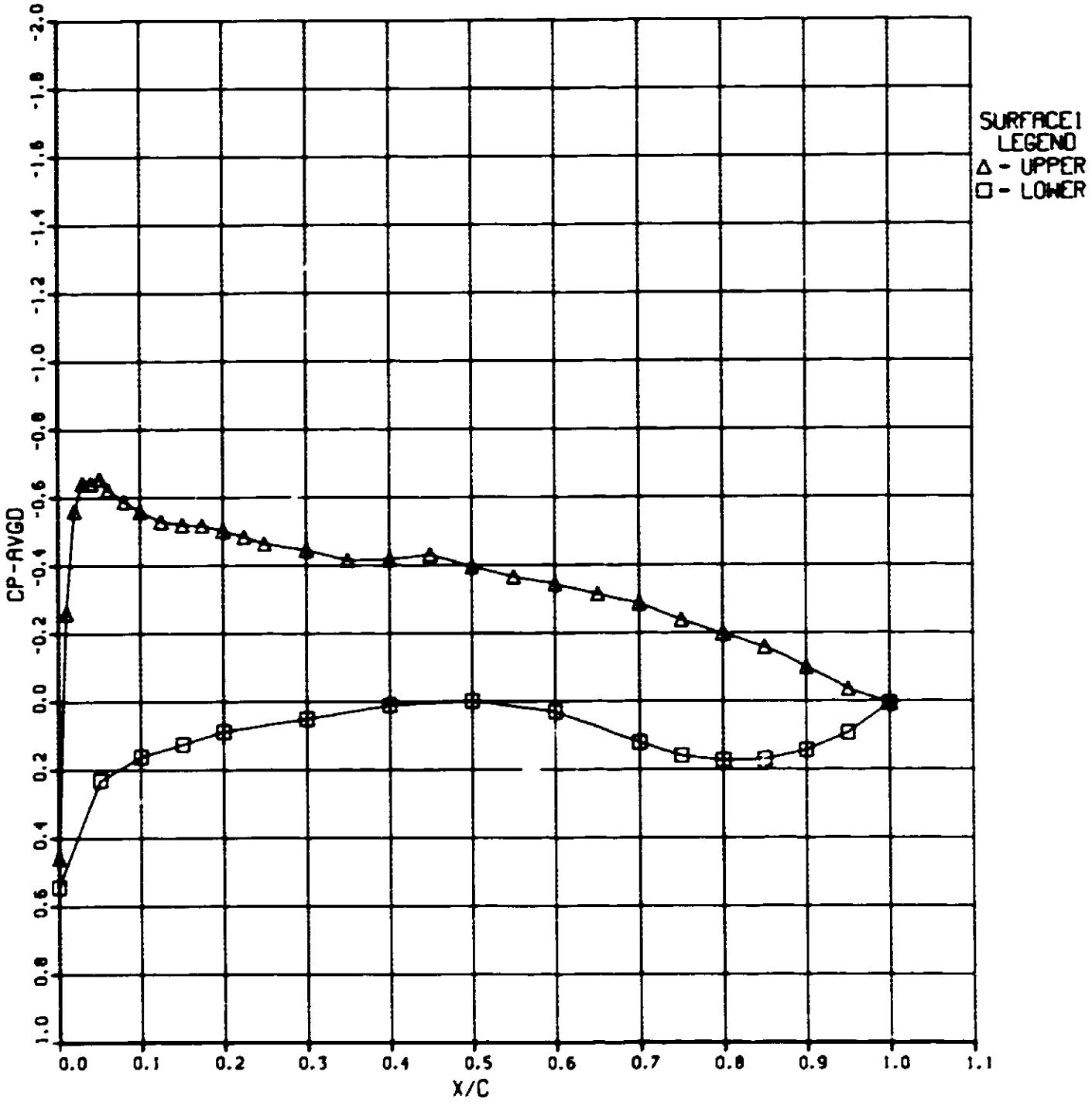

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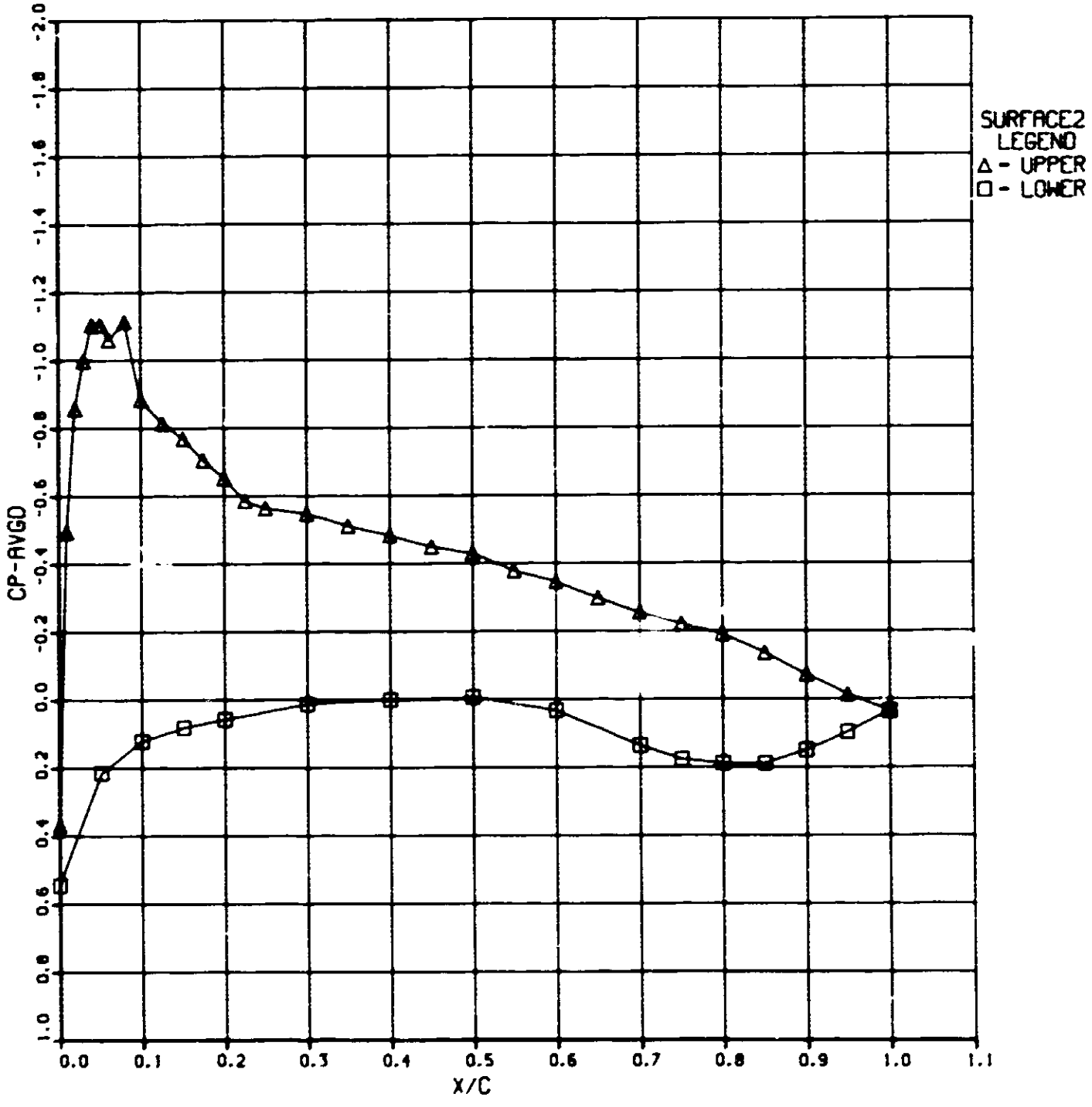

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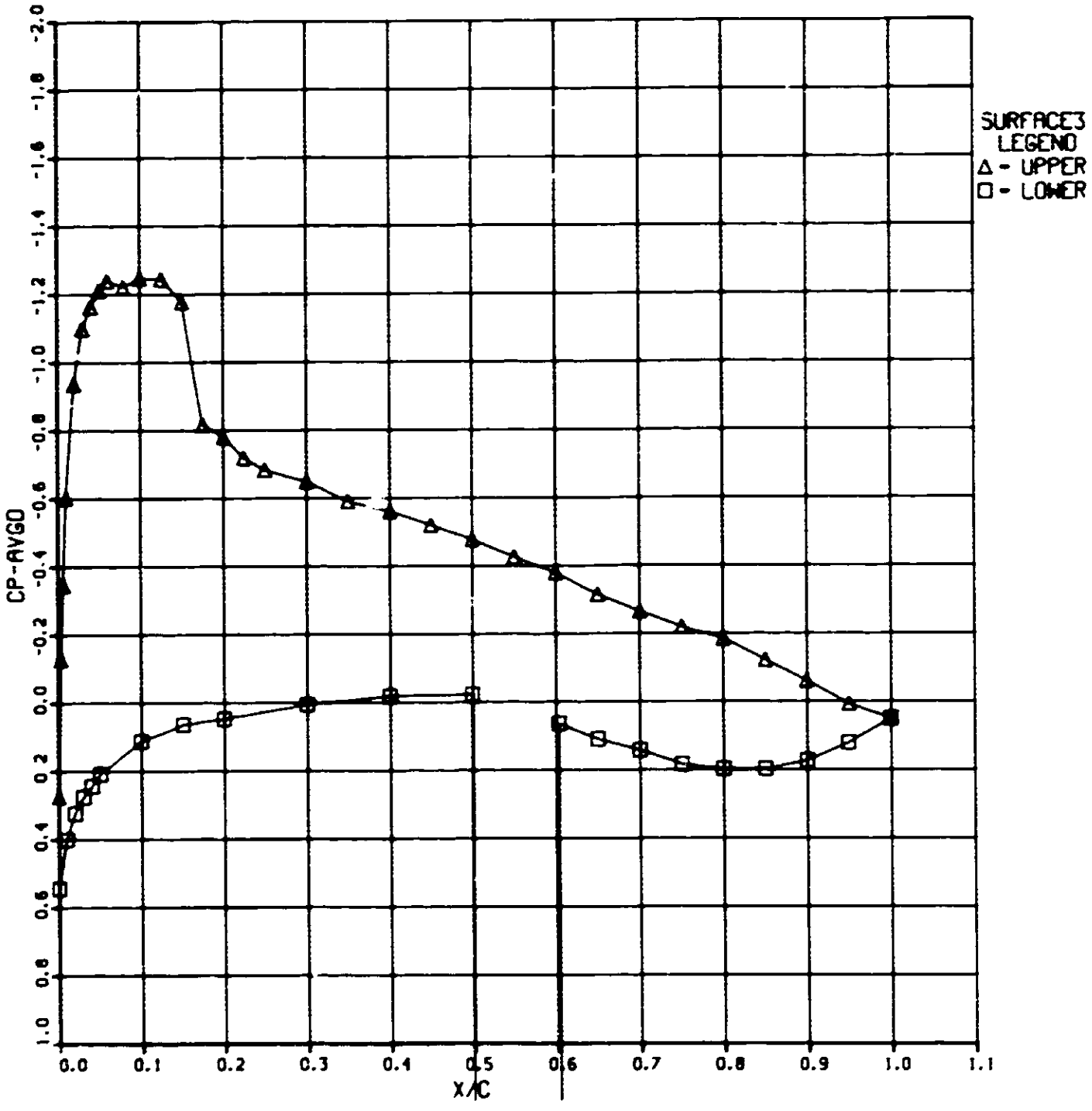

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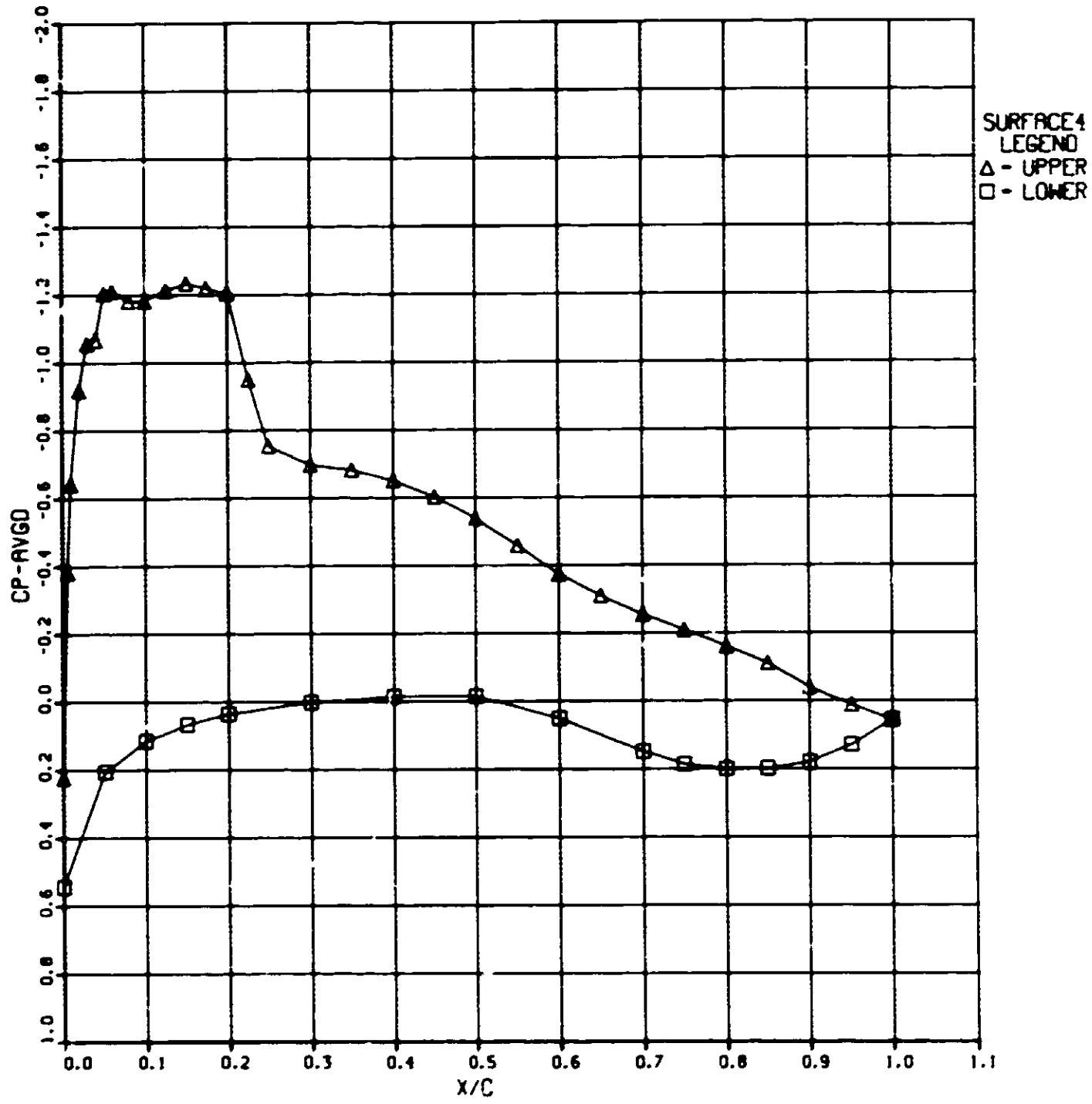

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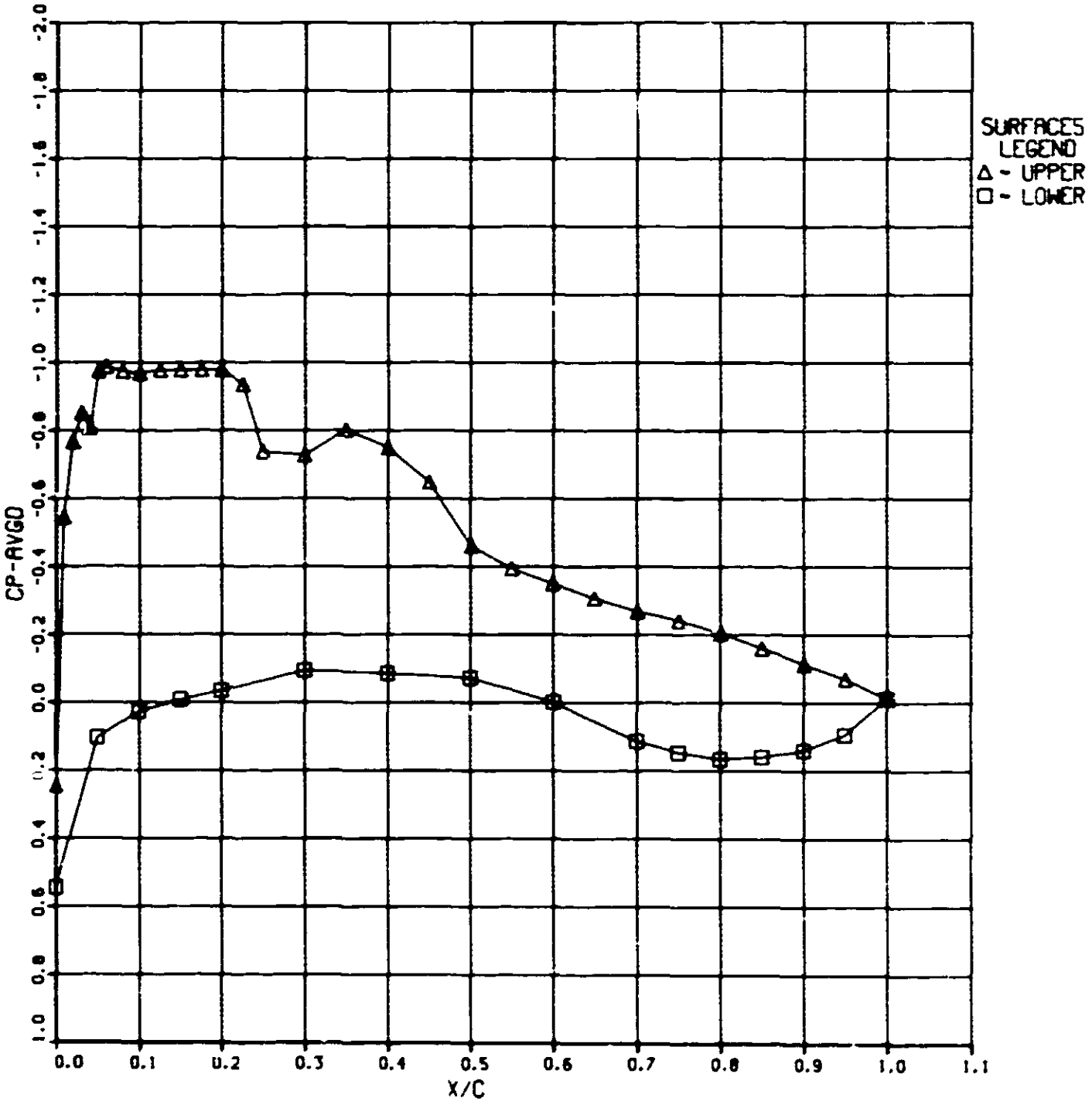

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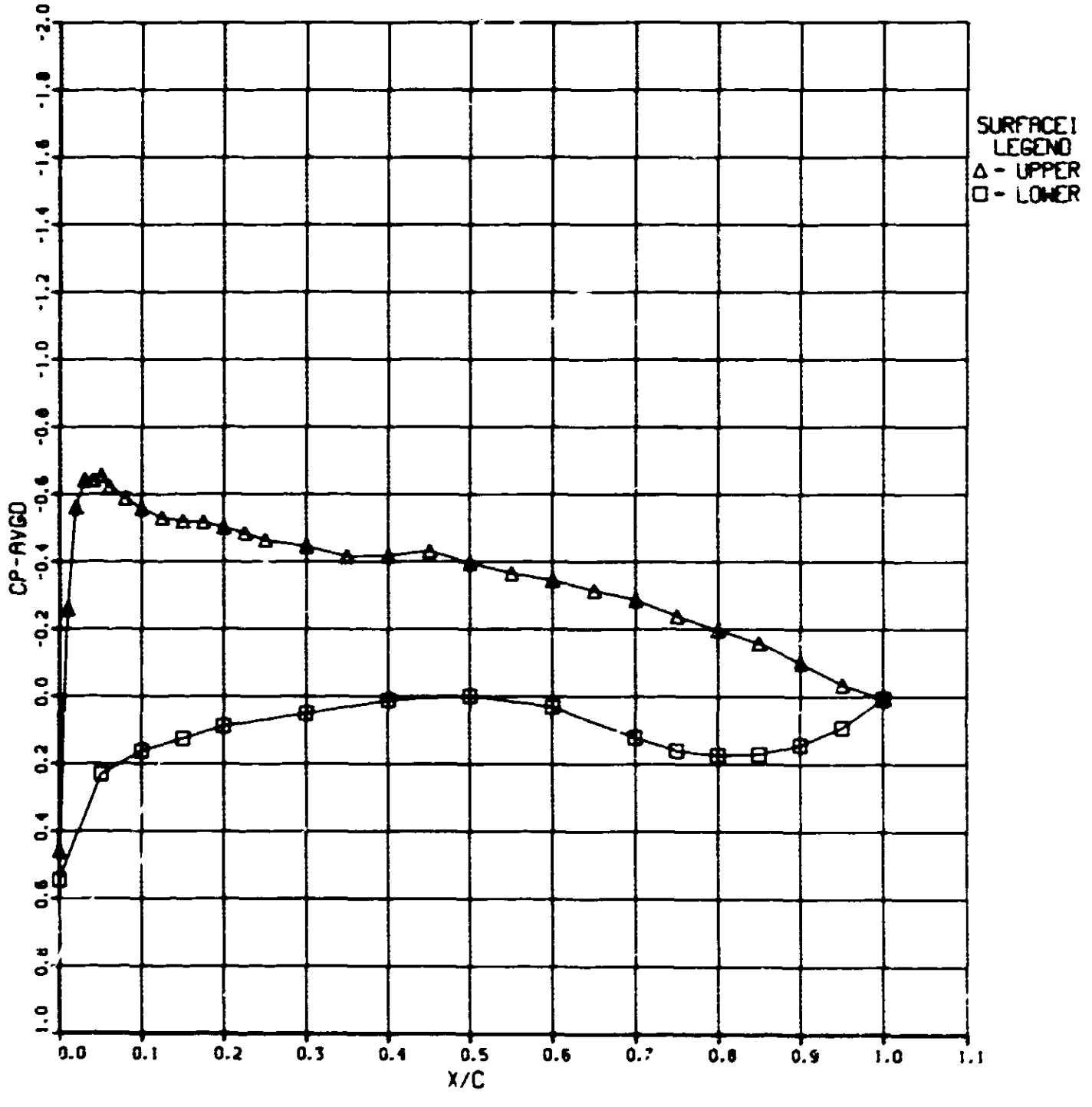

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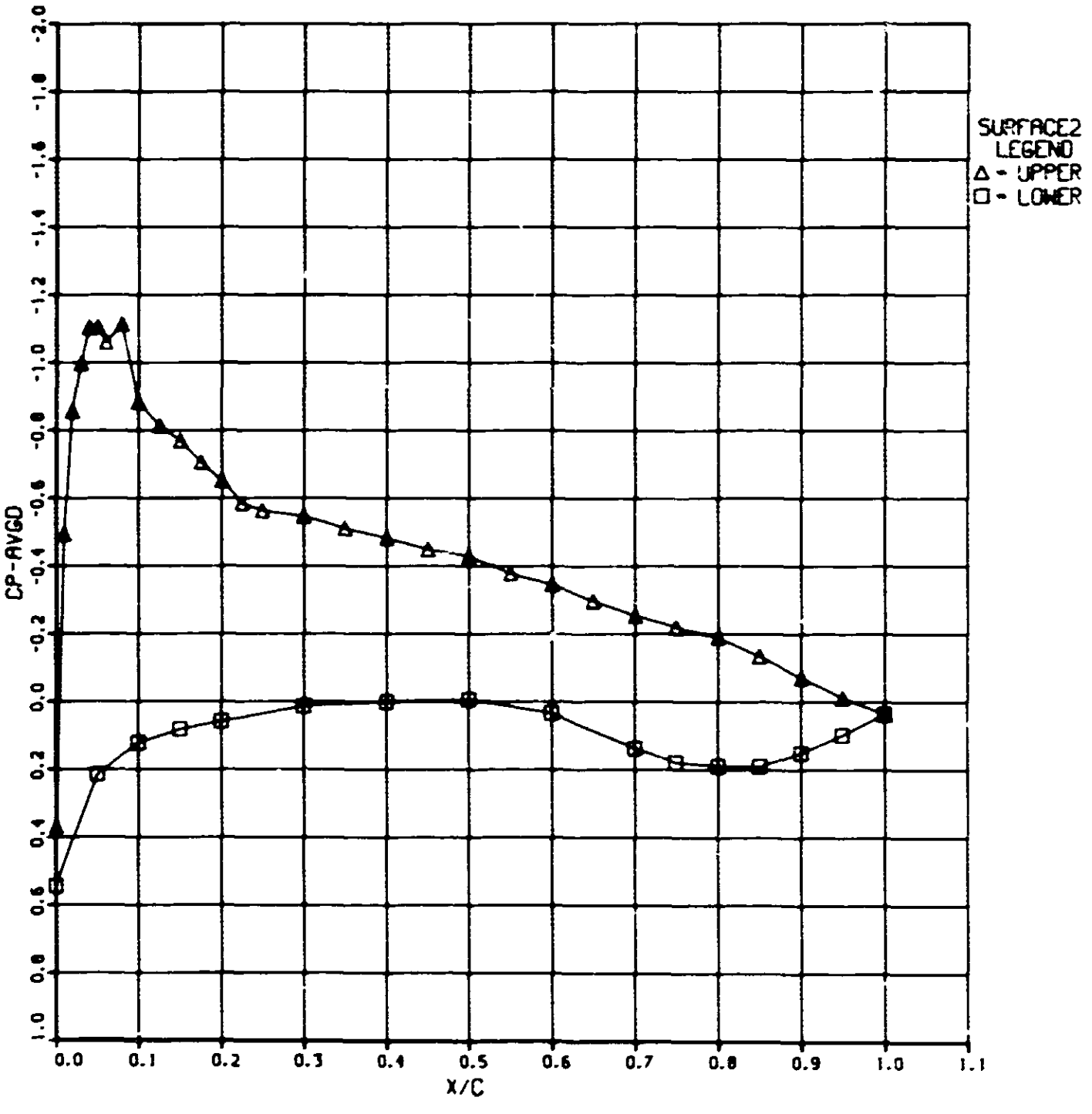

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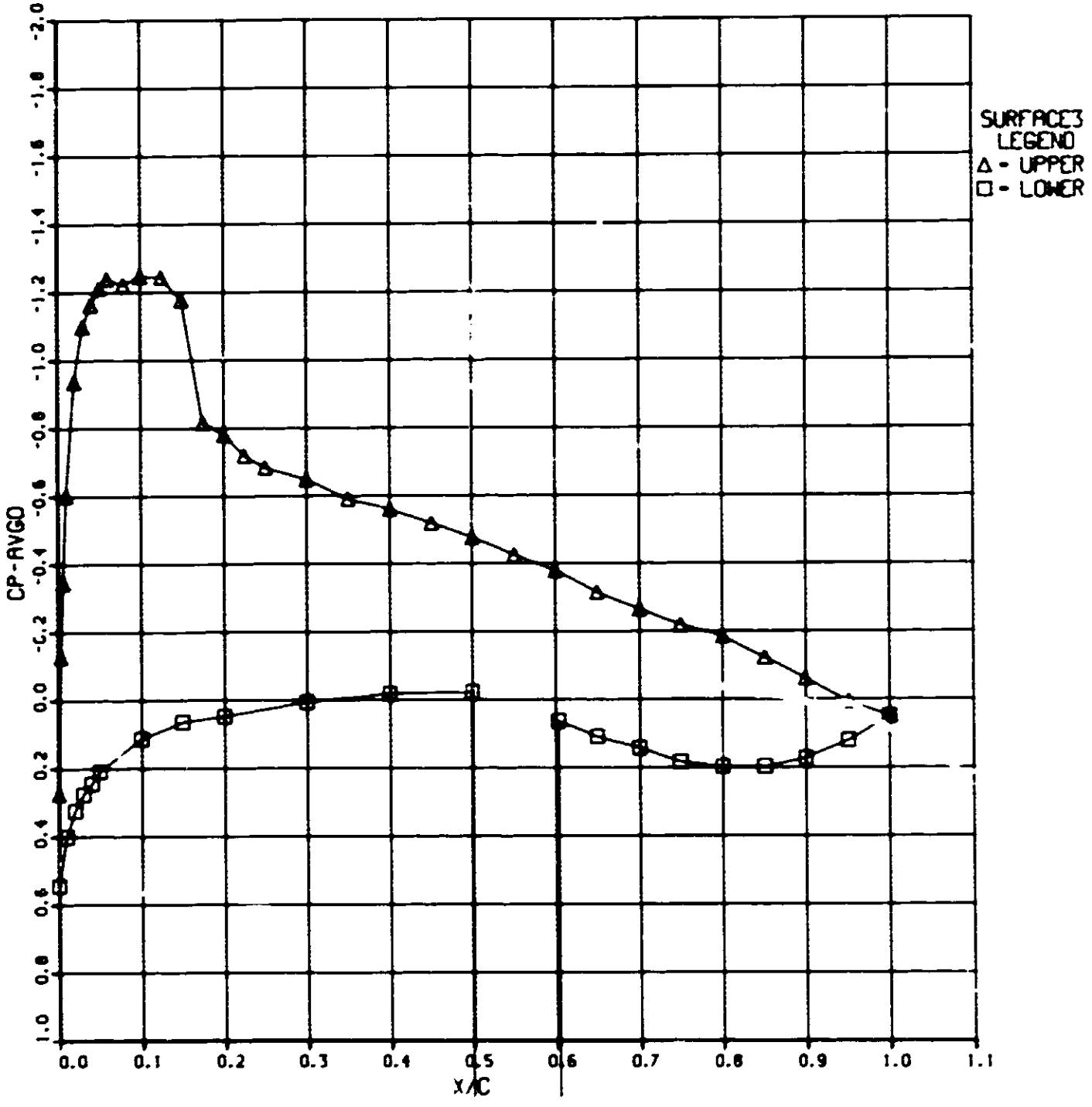

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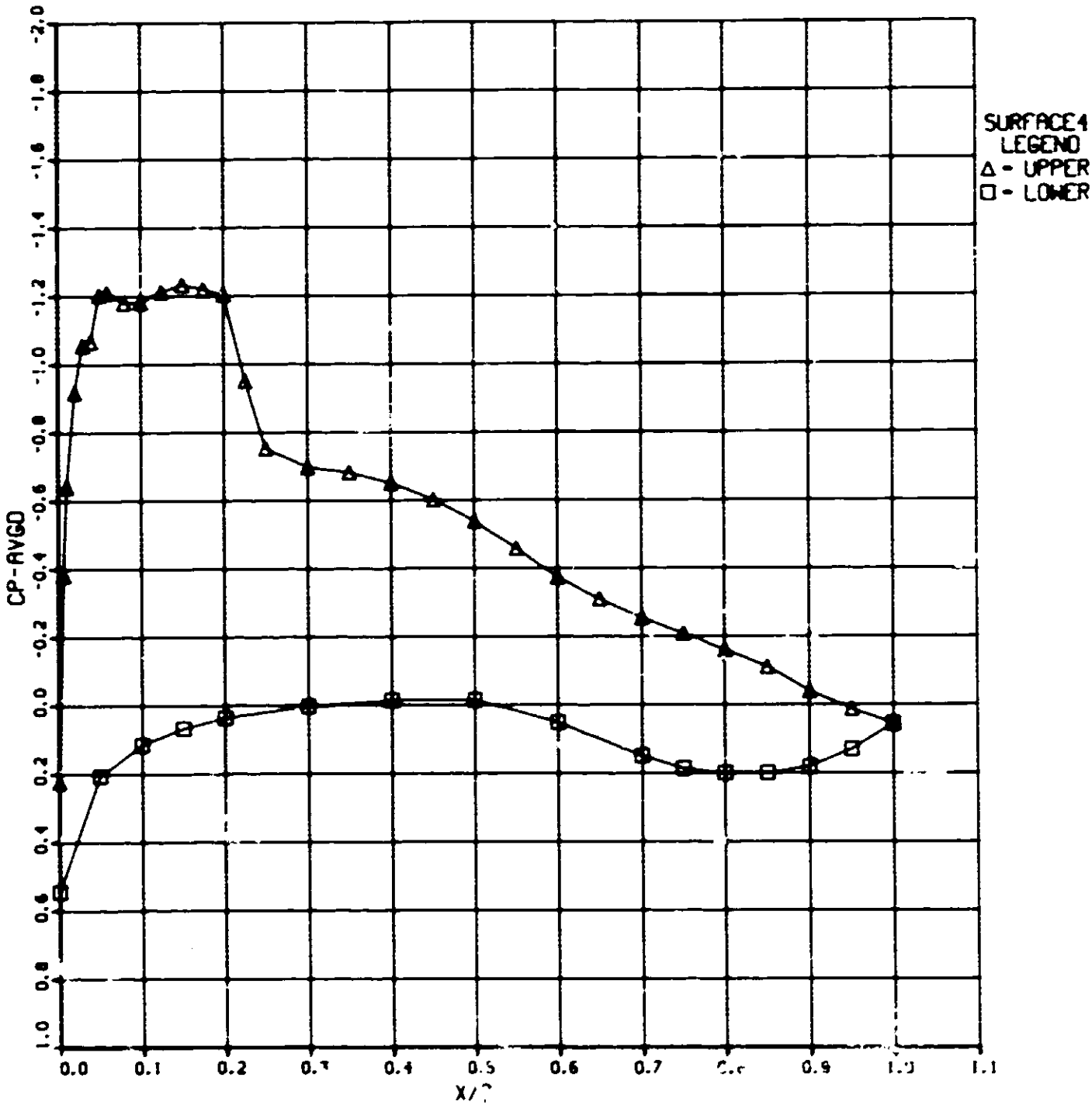

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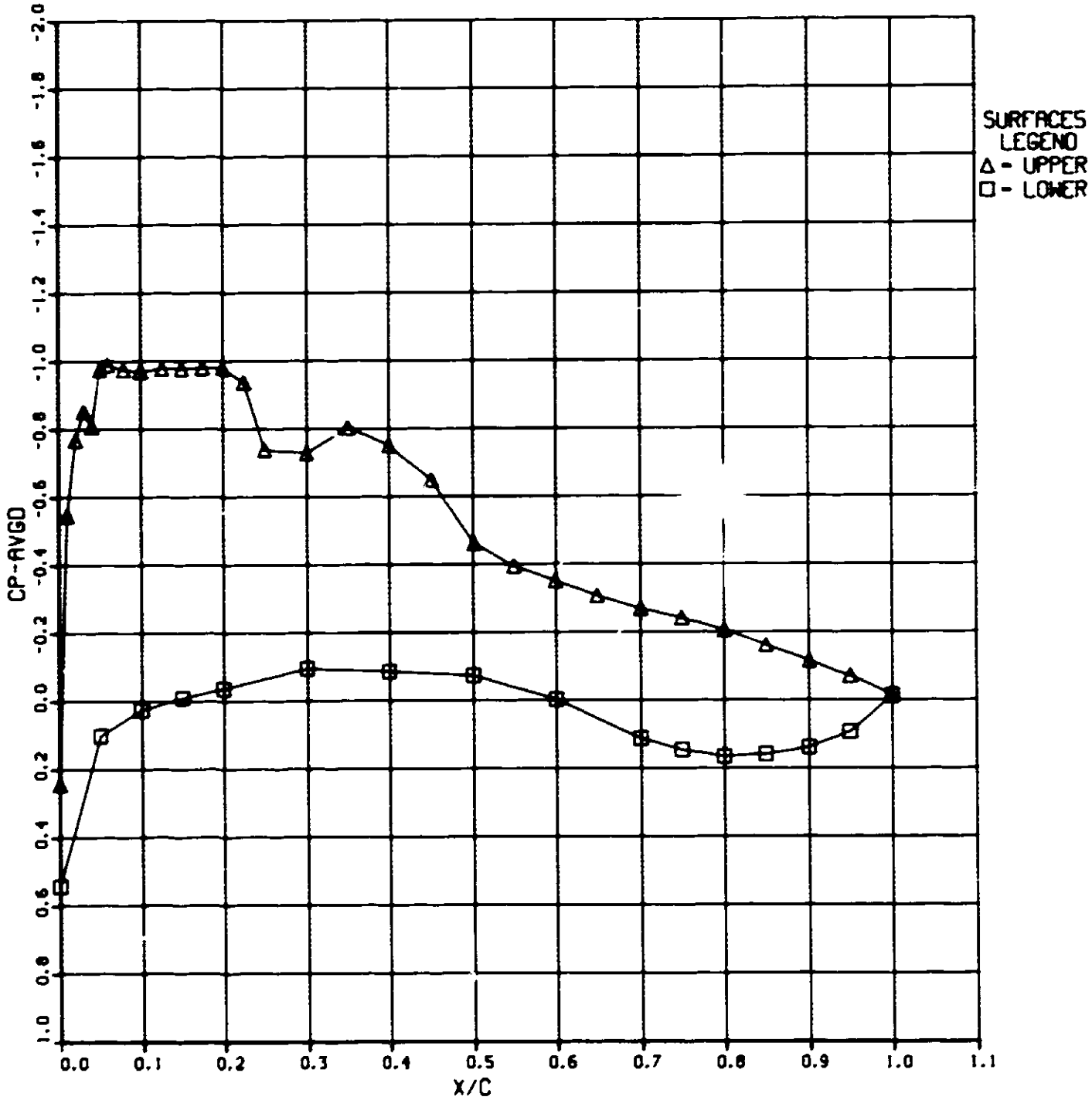
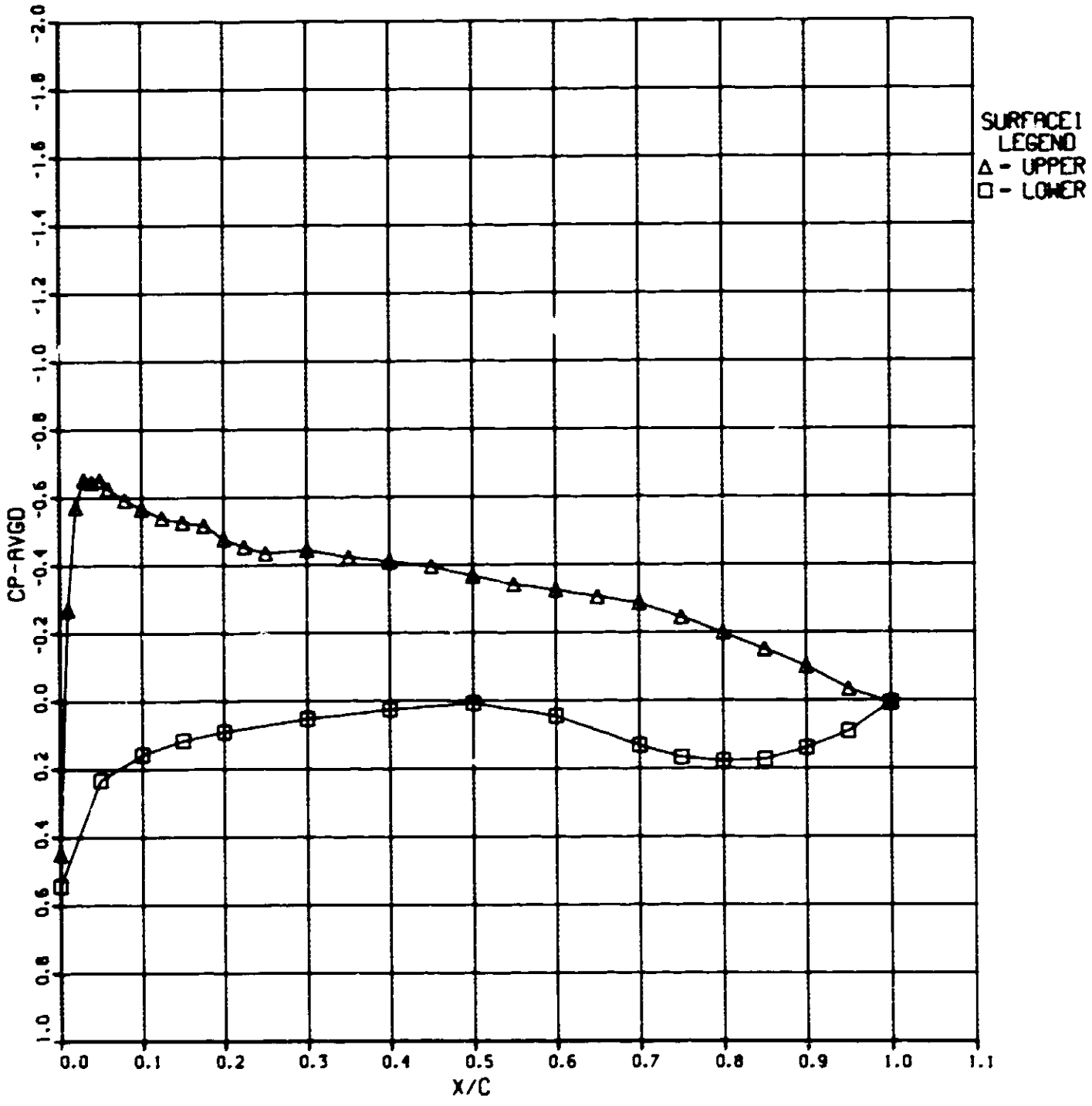


356-1-66 173.00: 2.00 CONF-17 MACH-0.812 RN-4.384 PT-2317 ALPHA- 5.00



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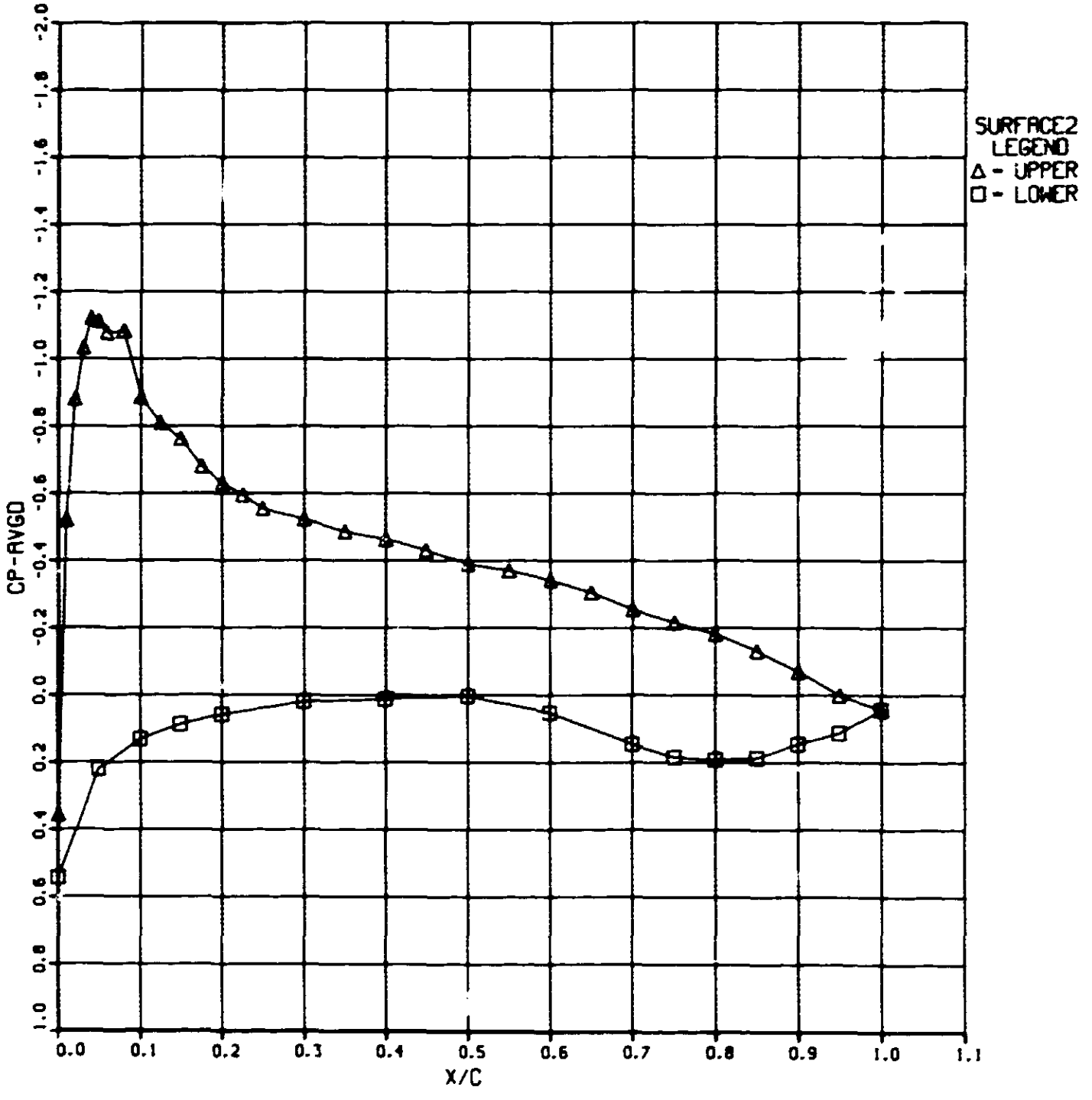

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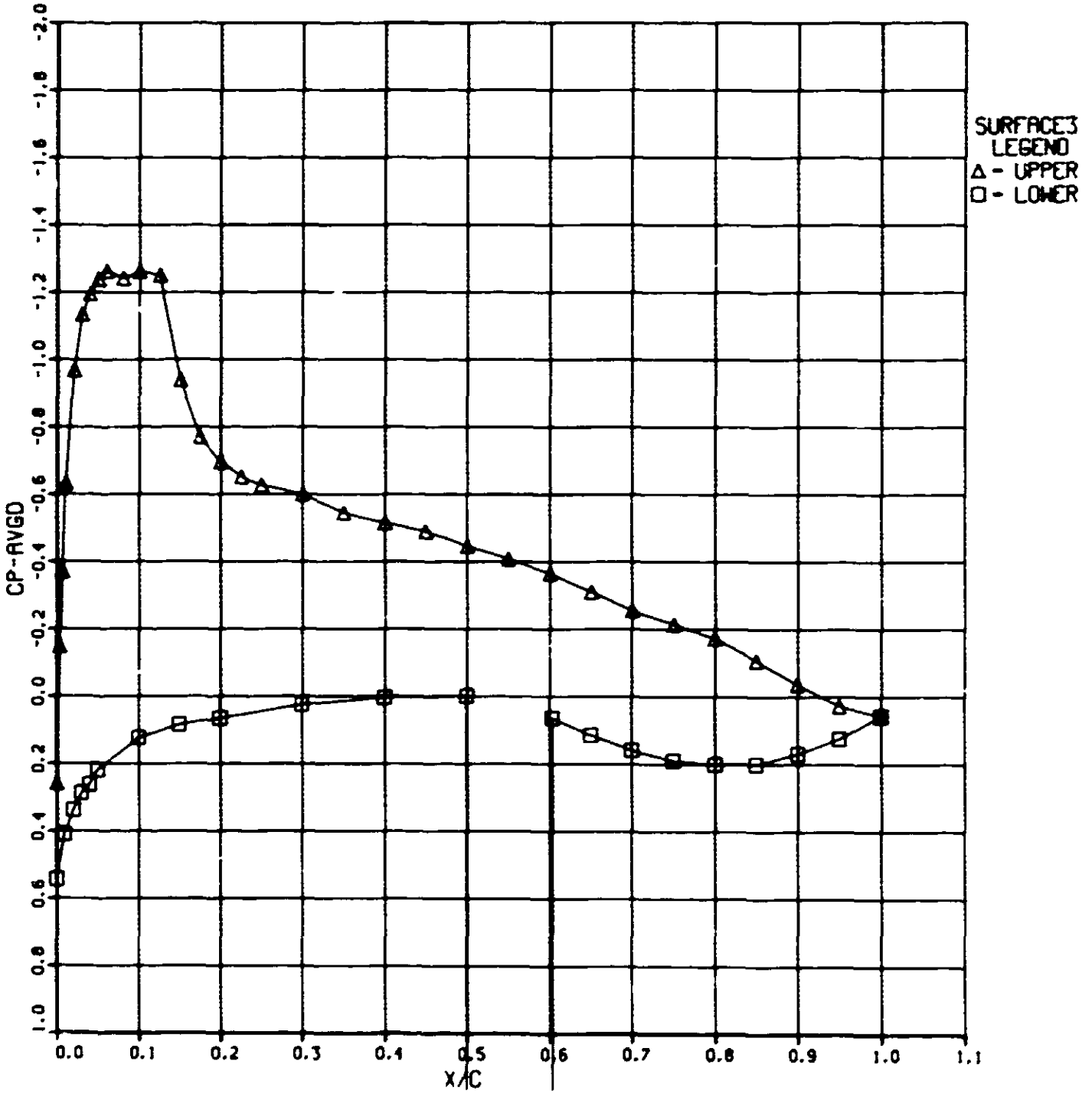

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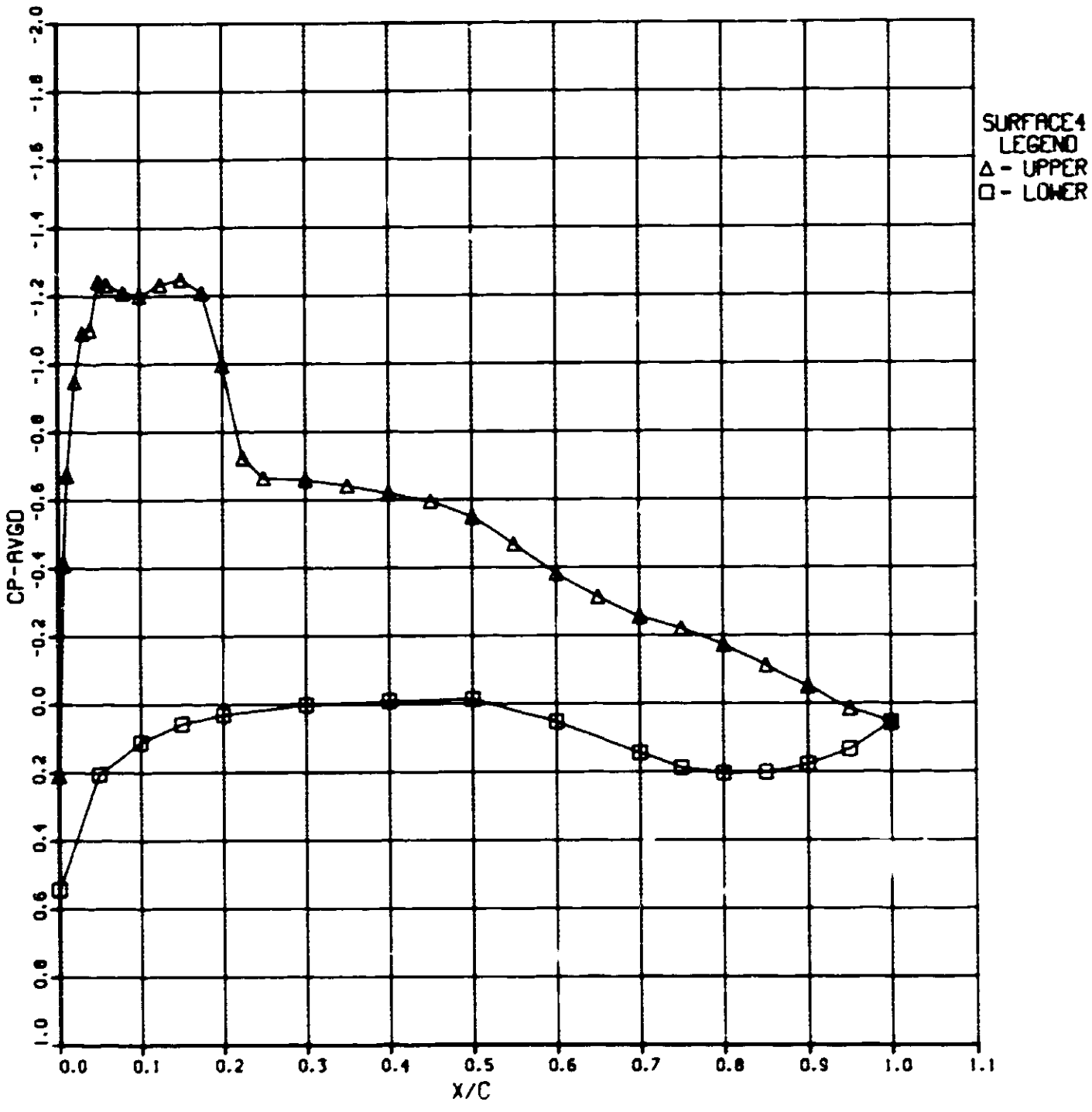

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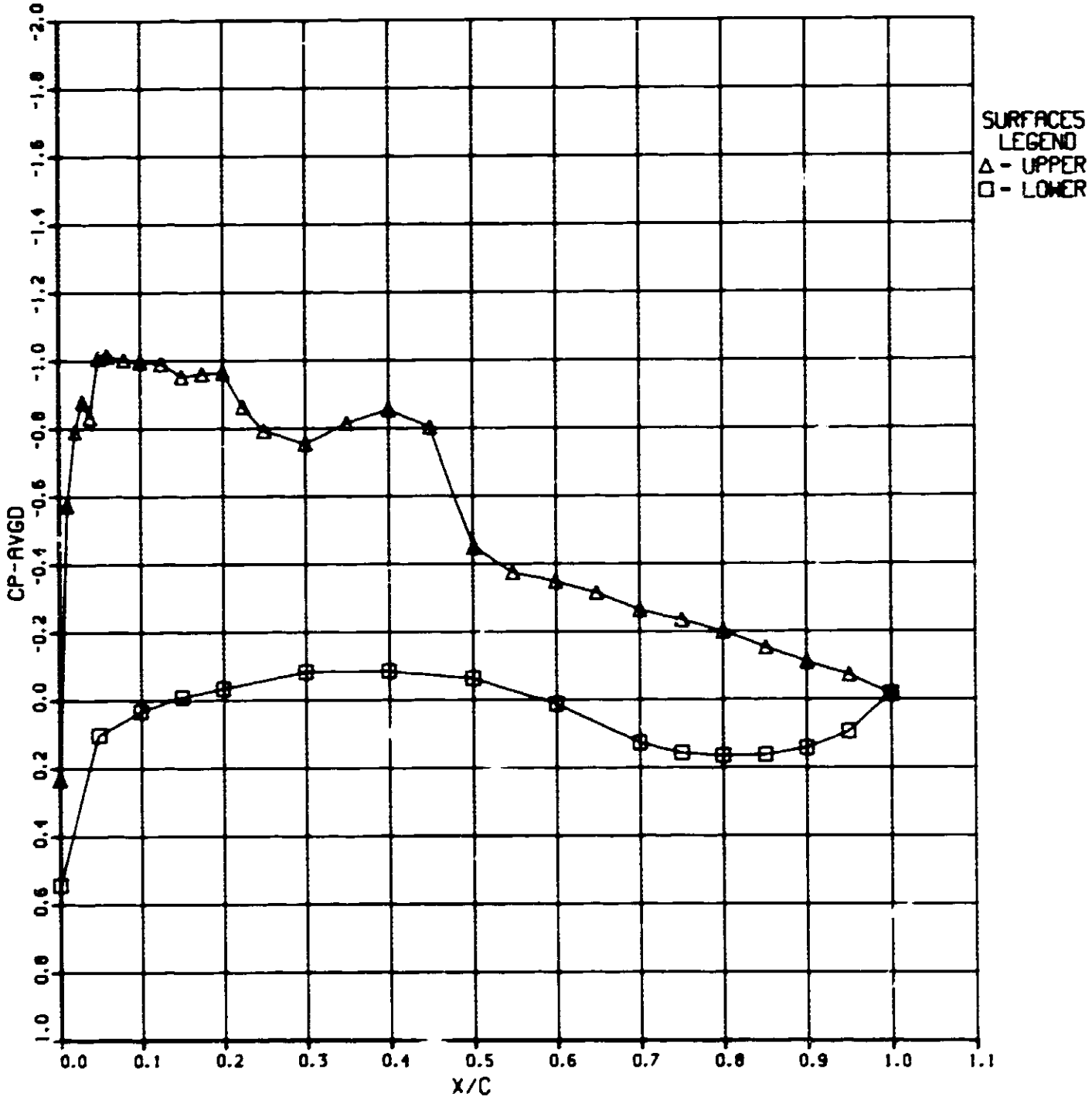
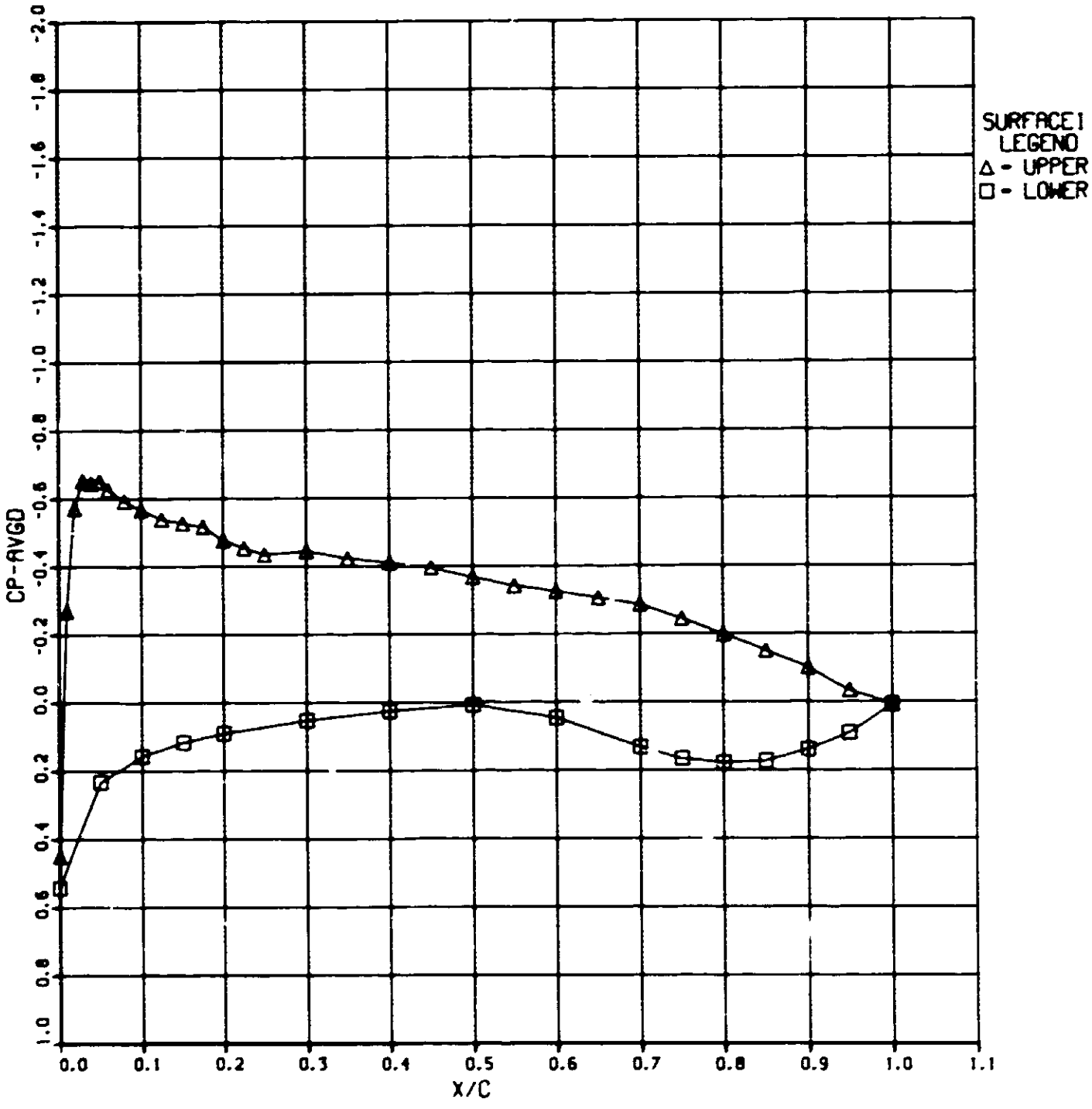


356-1-66 173.00: 2.00 CONF-17 MACH-0.812 RN-4.384 PT-2317 ALPHA- 5.00



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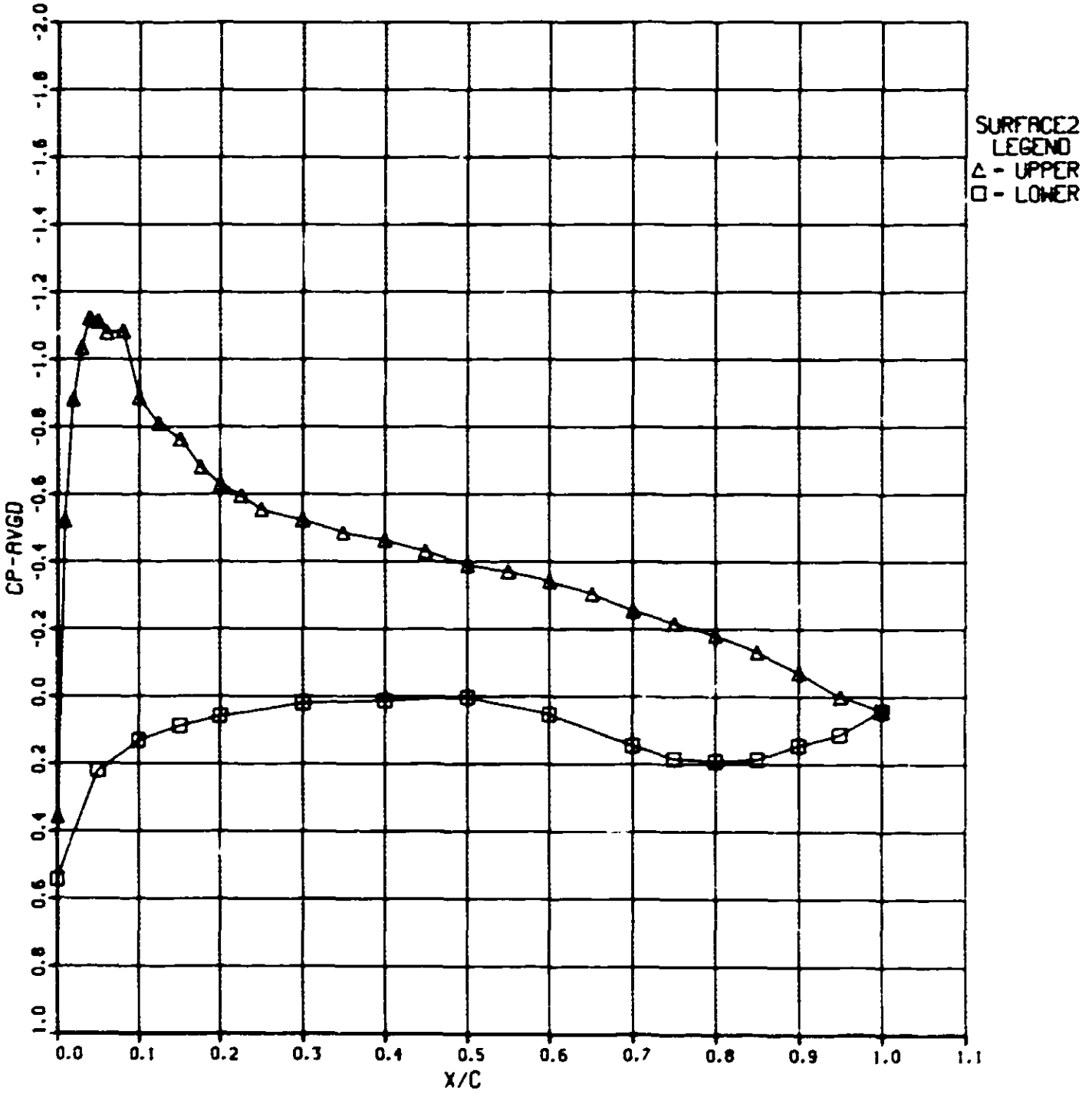

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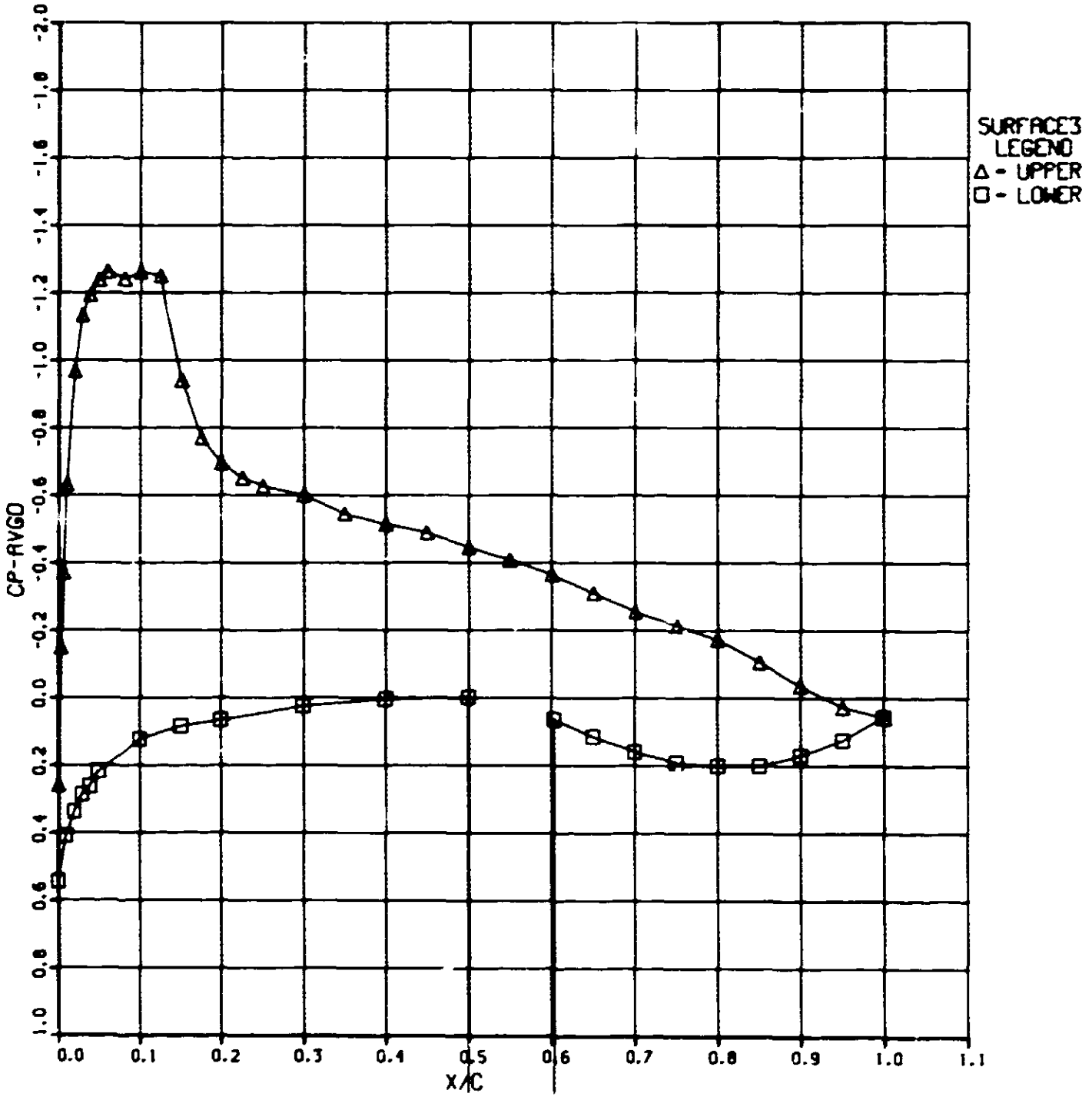

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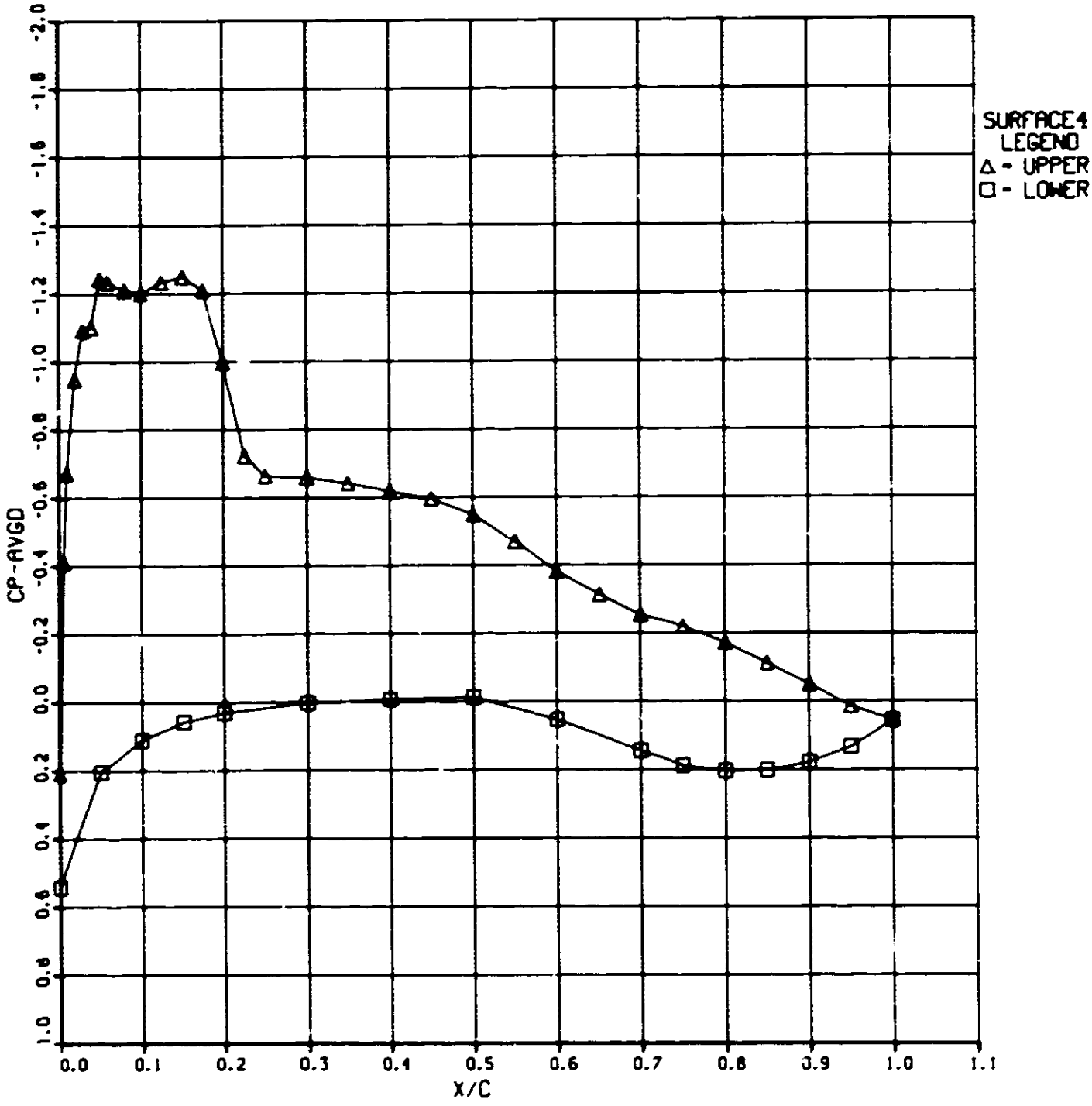

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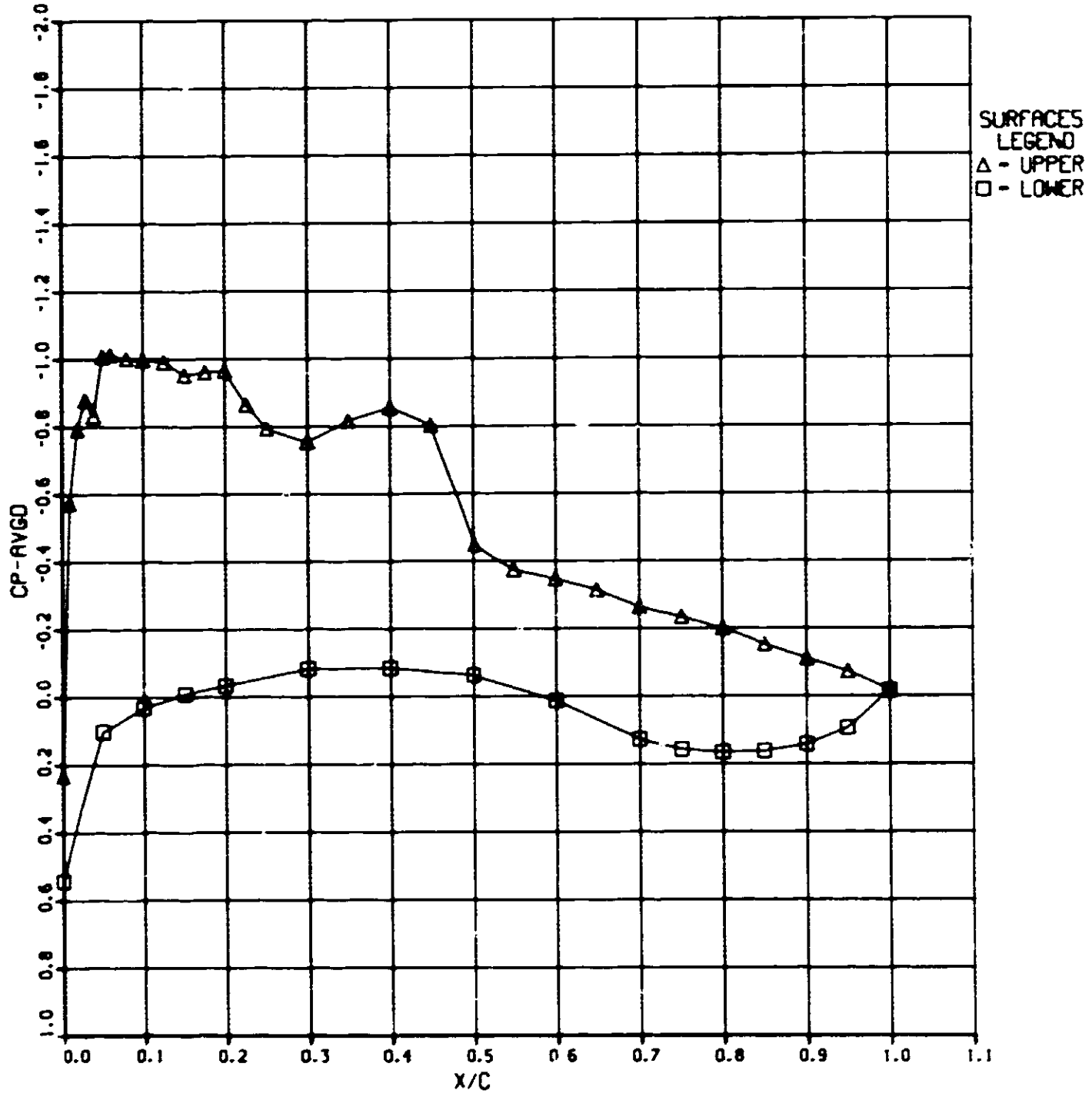

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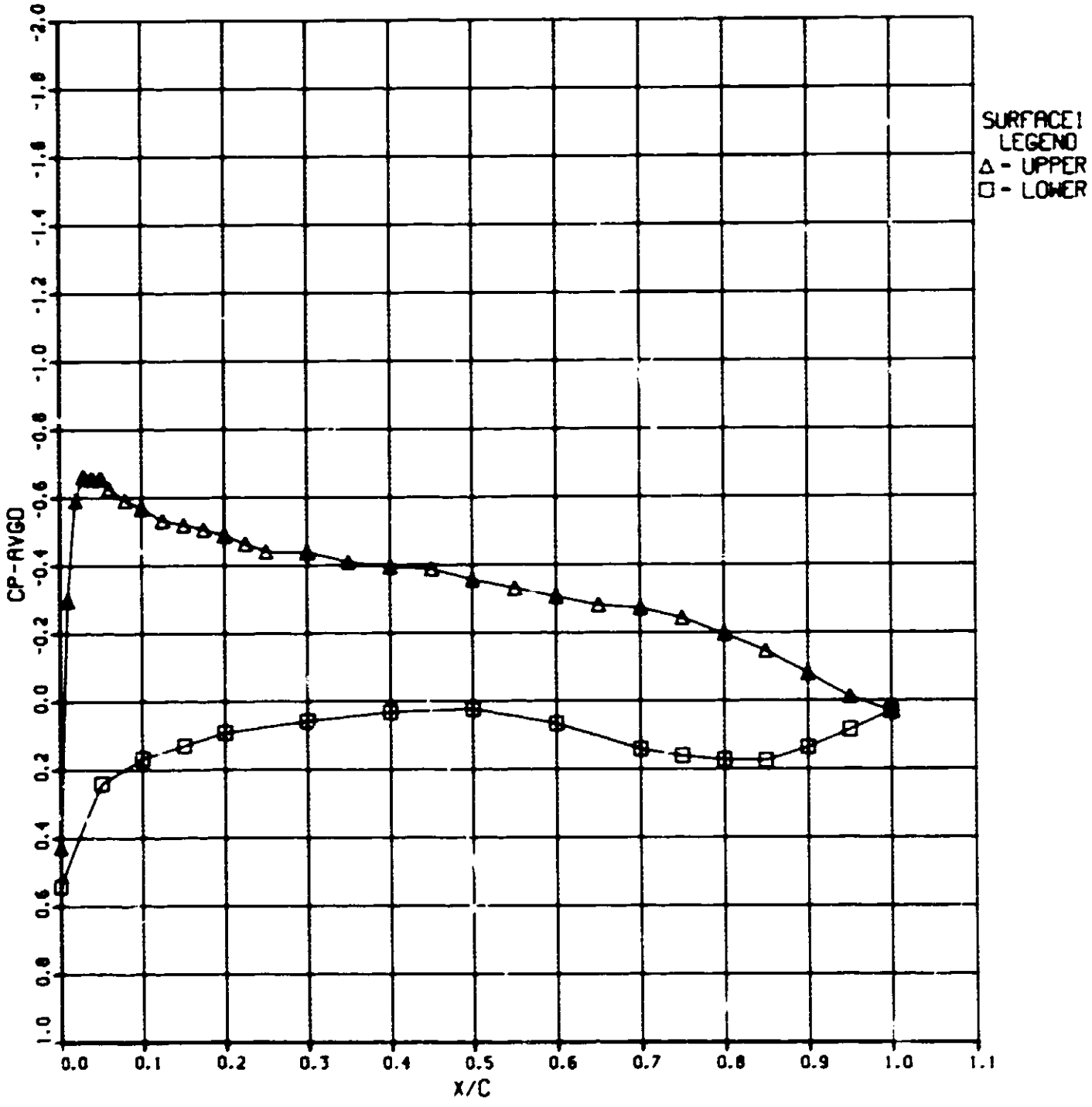

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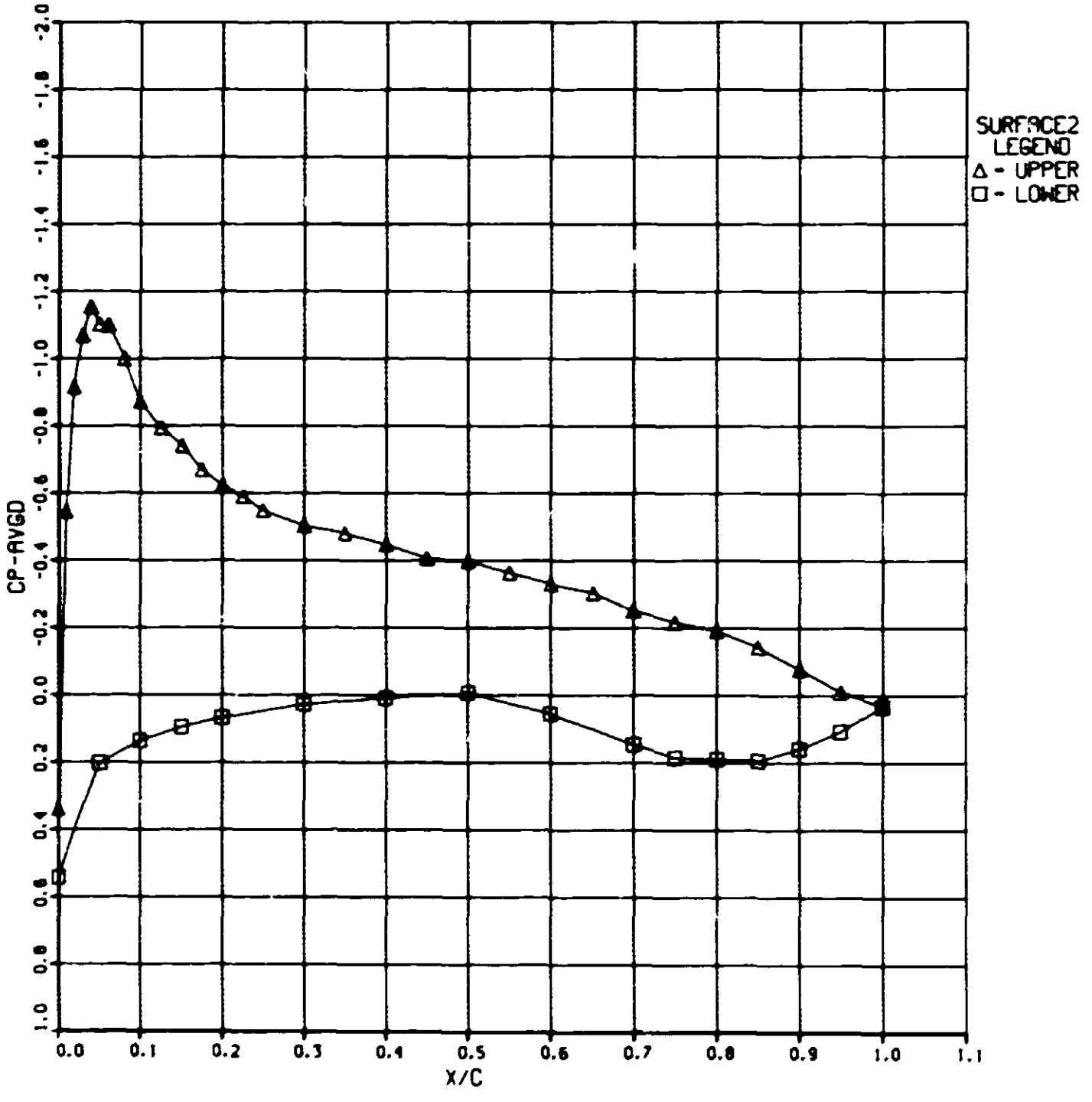

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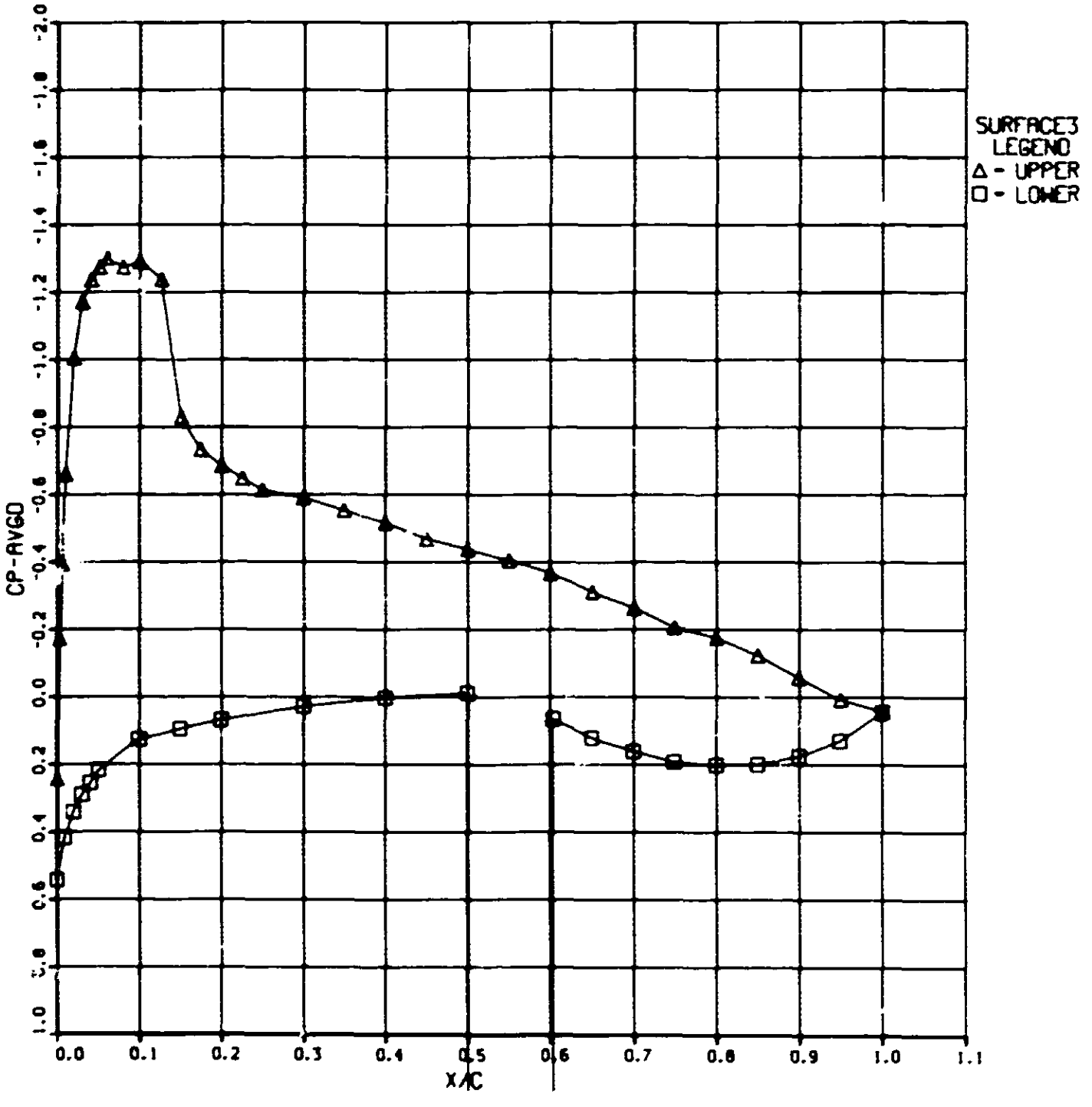

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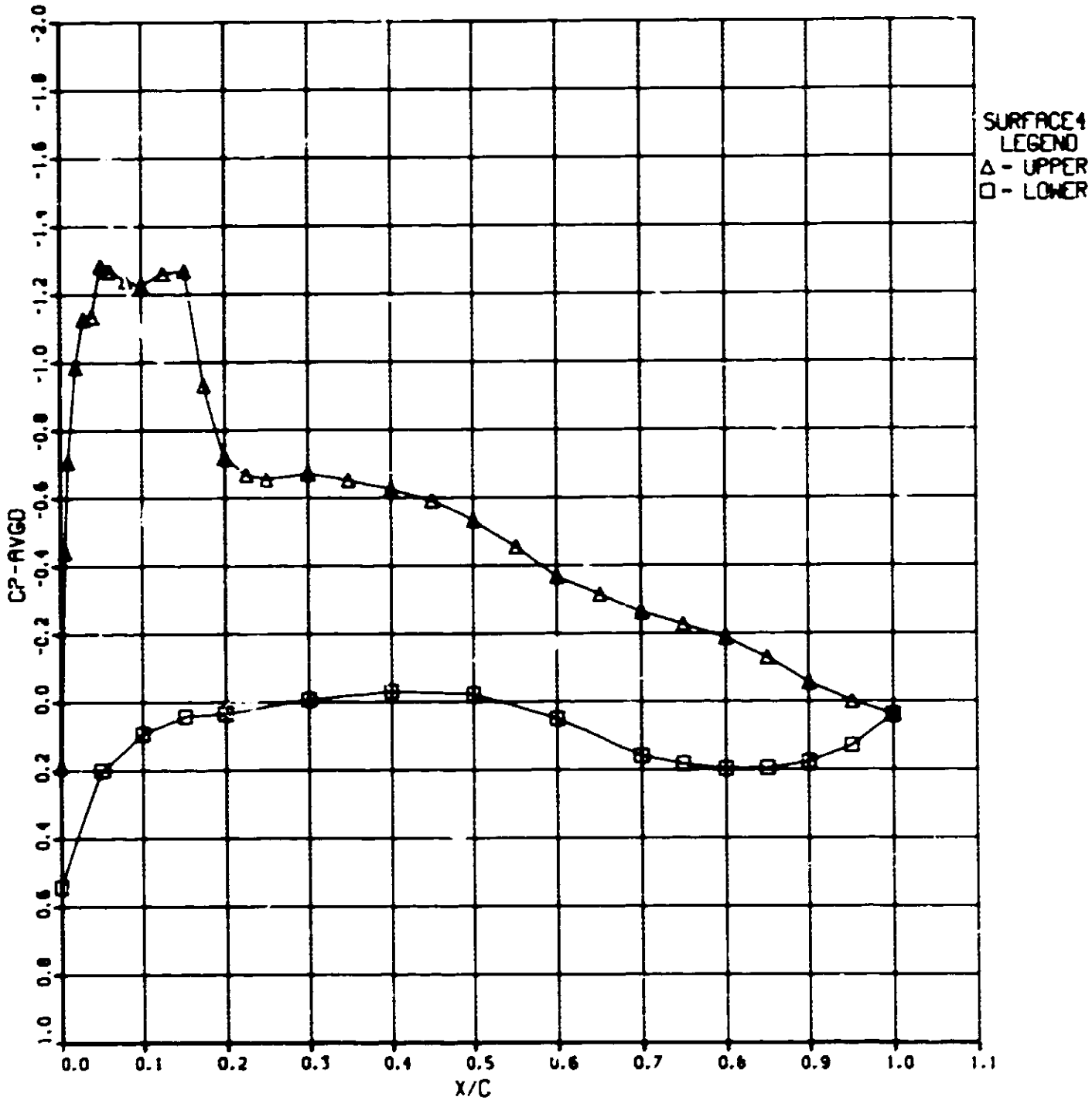

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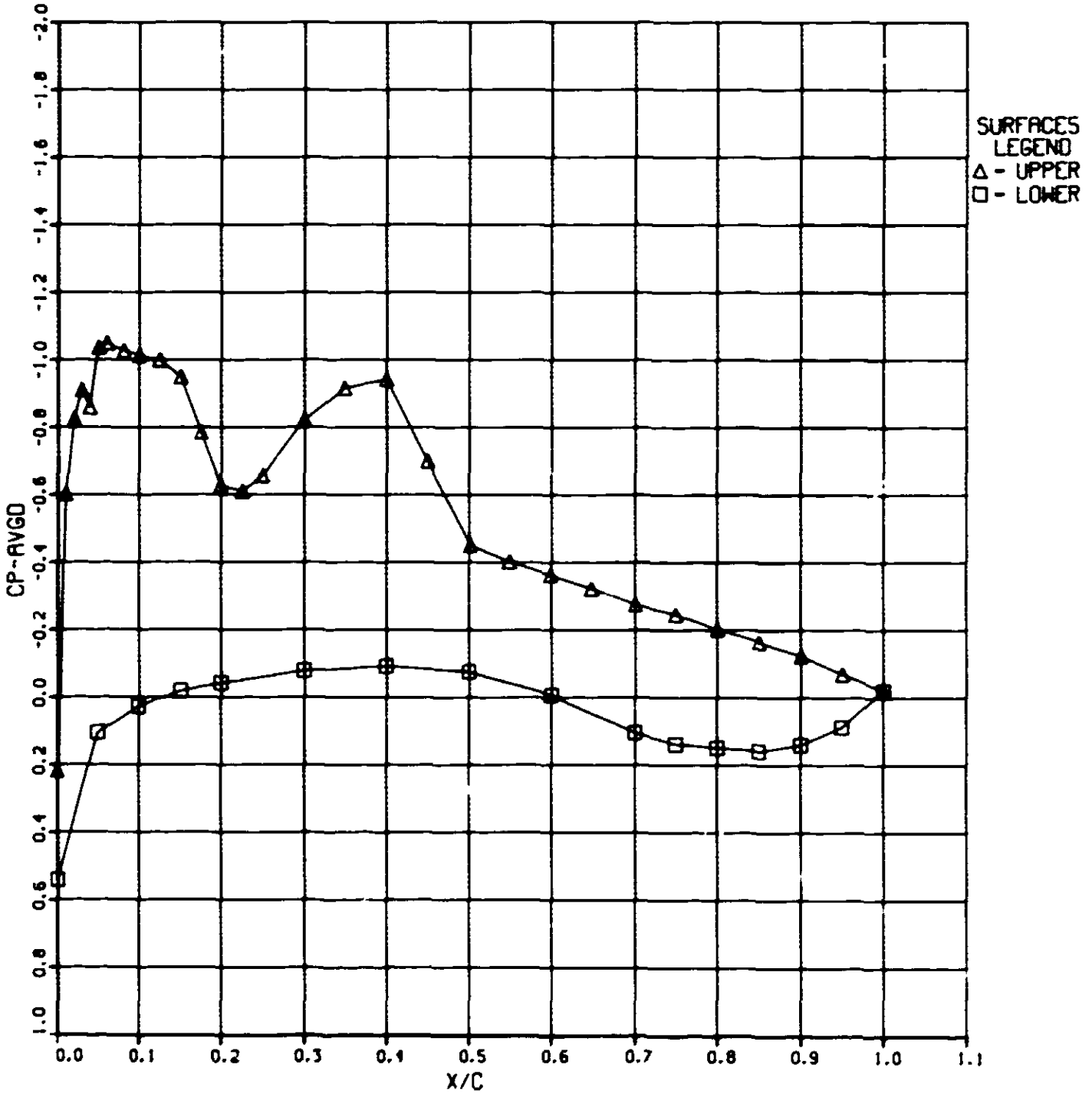

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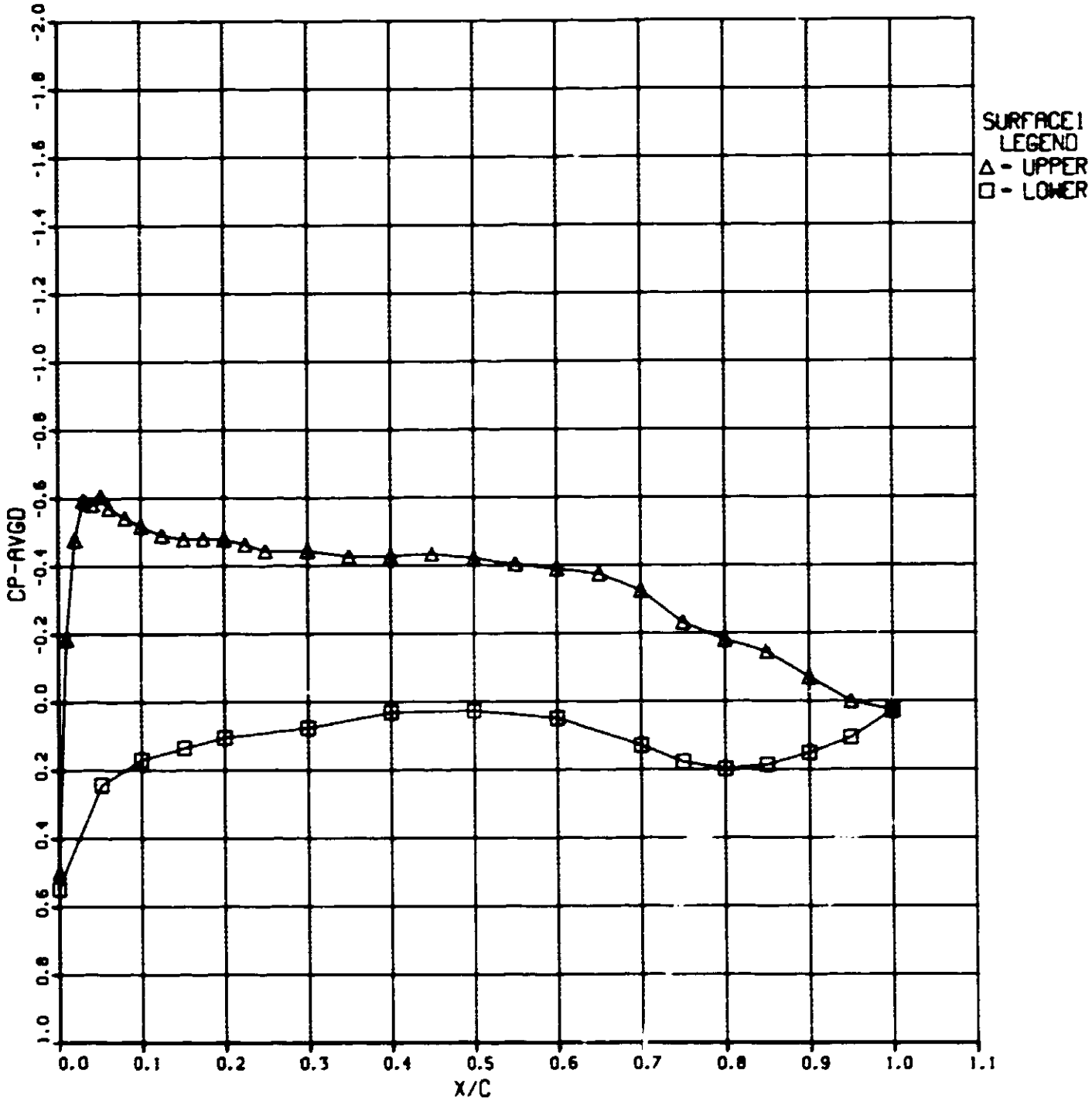

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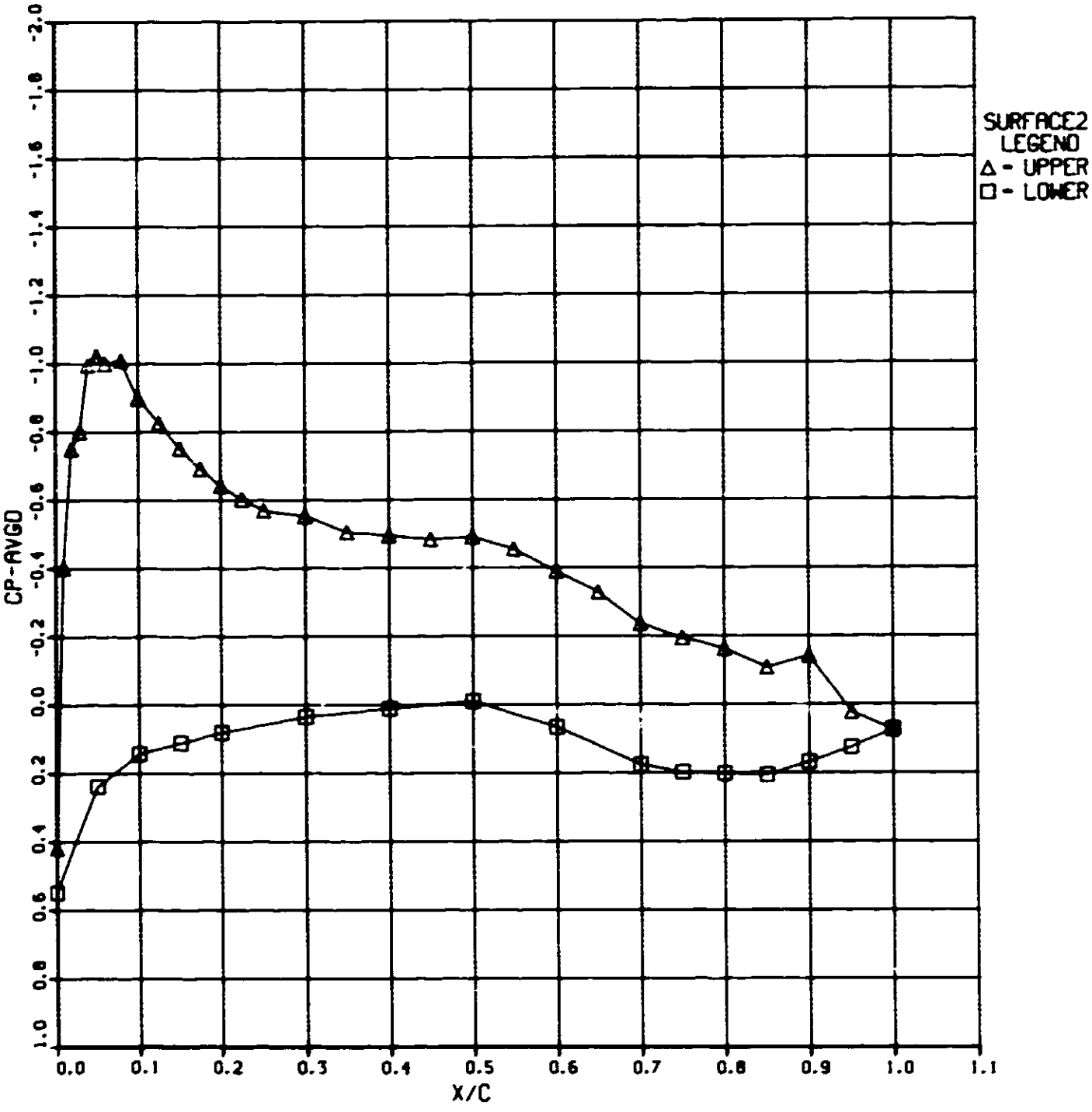

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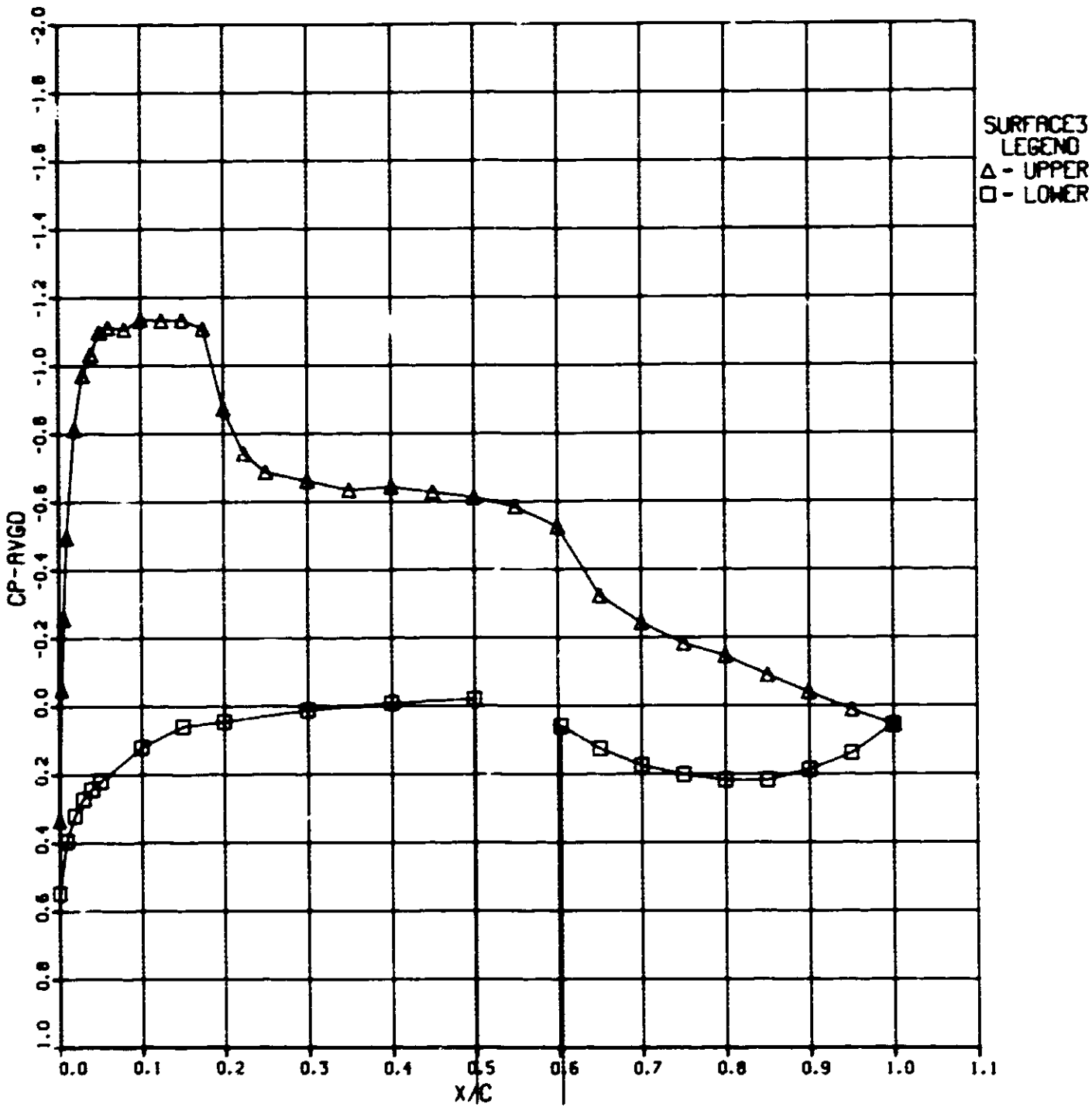

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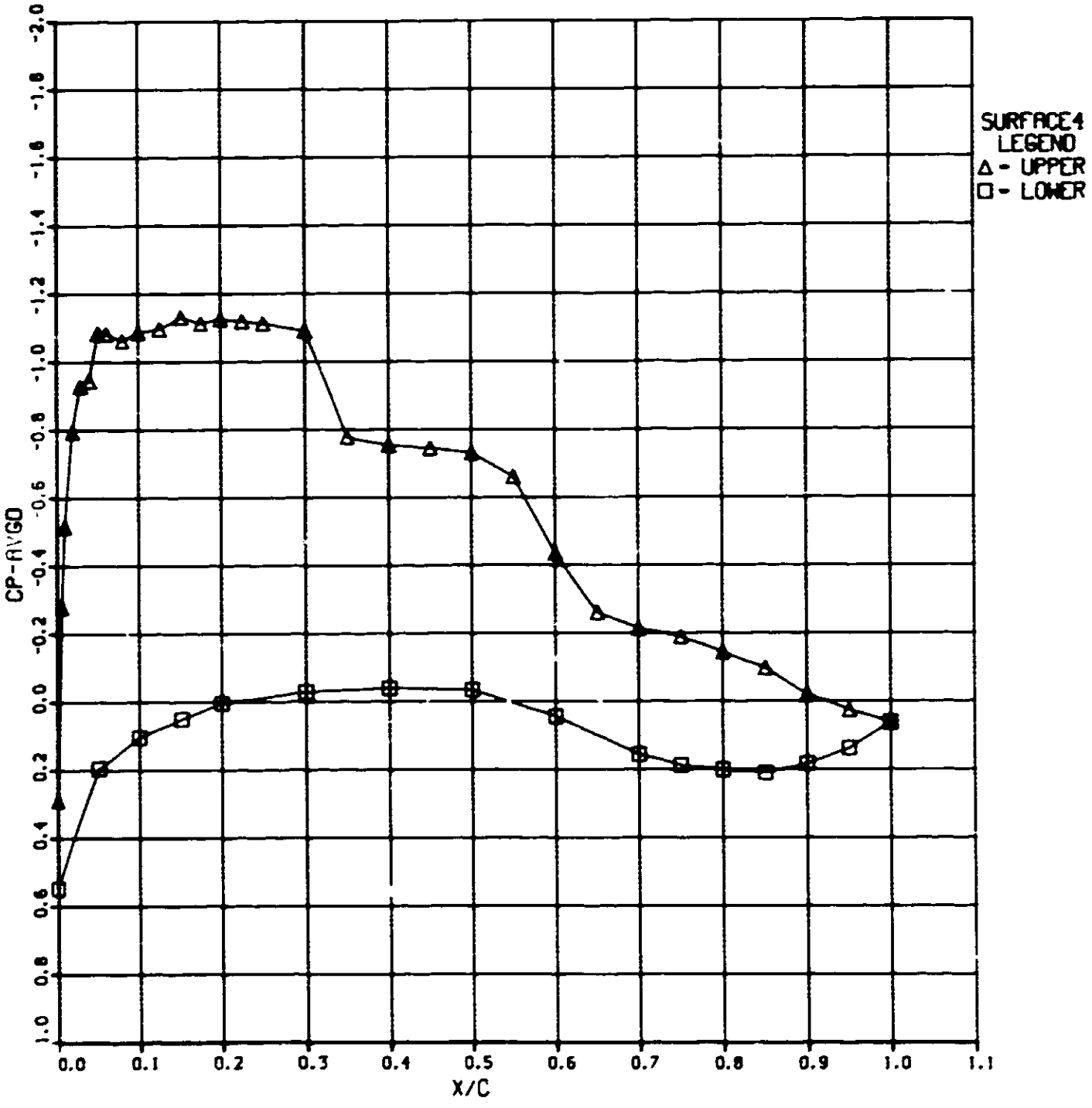

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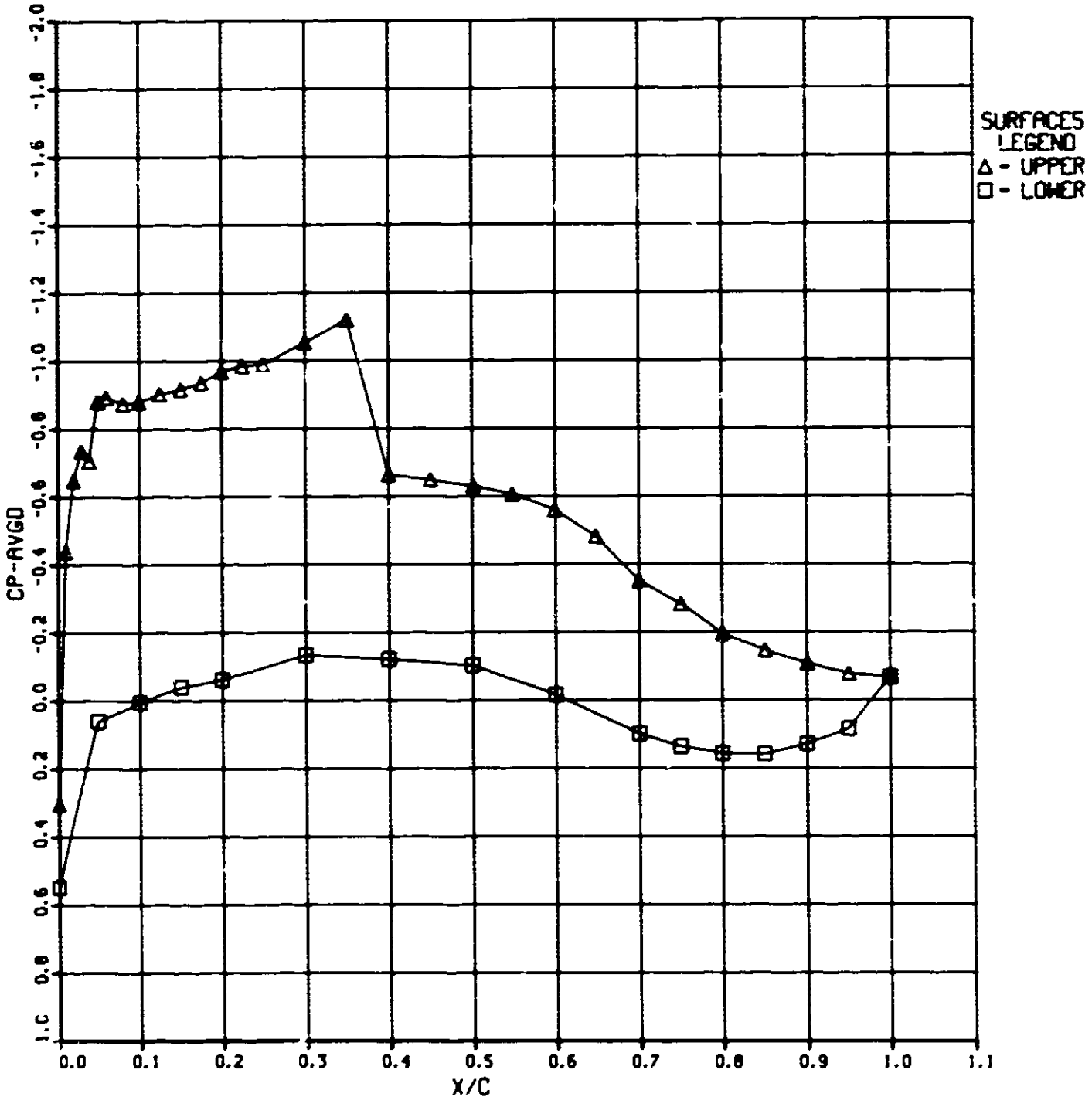

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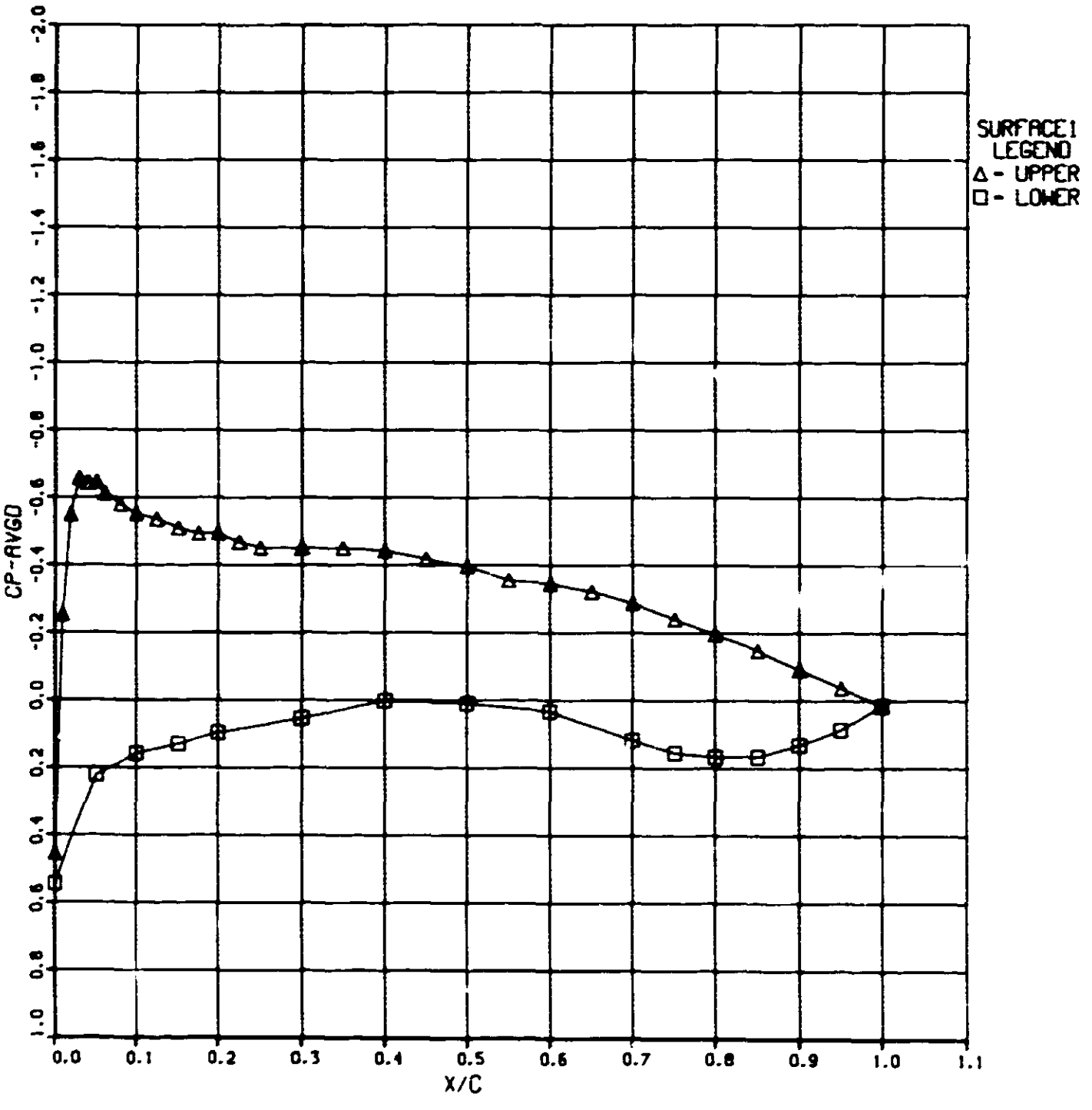

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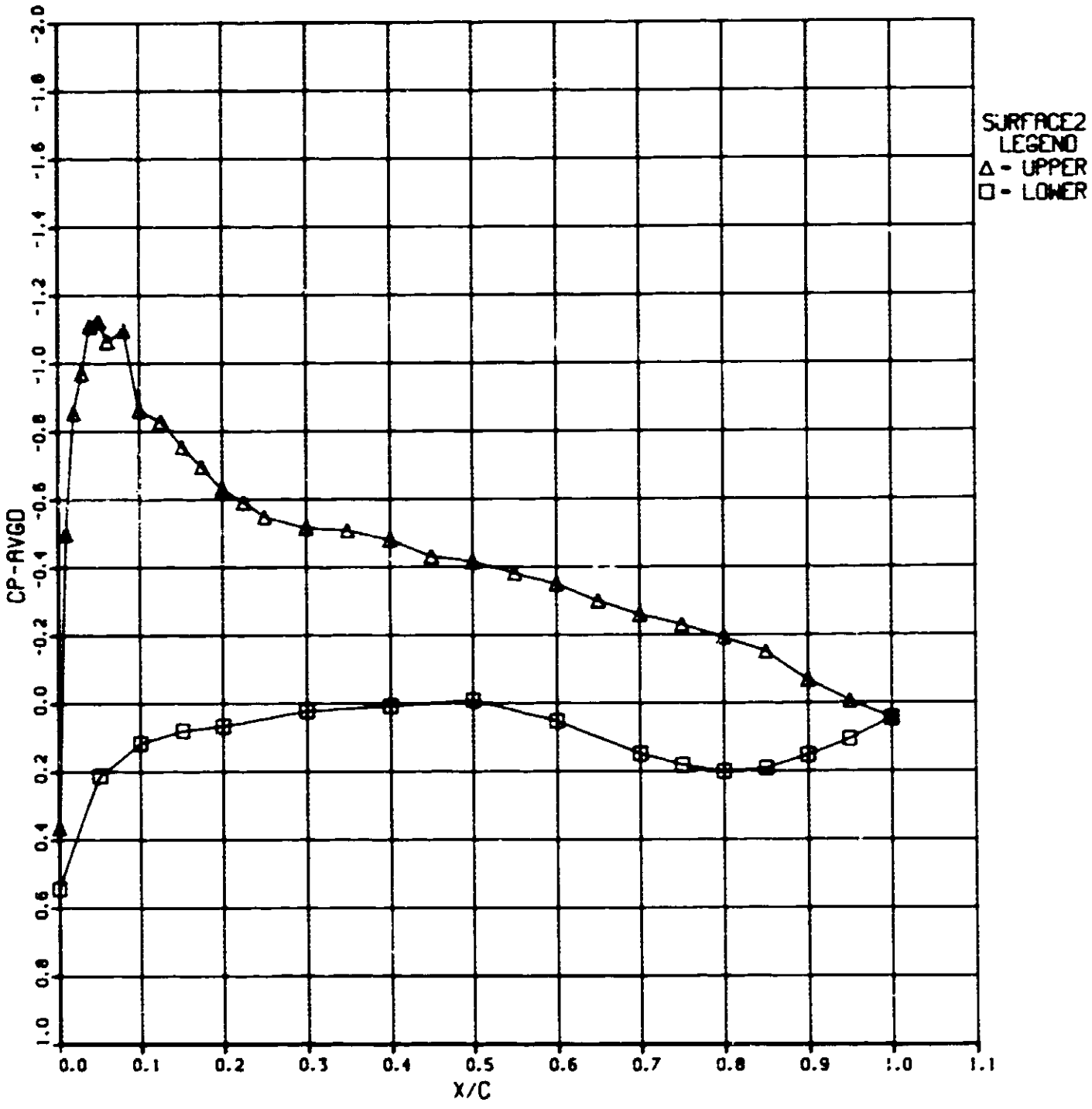

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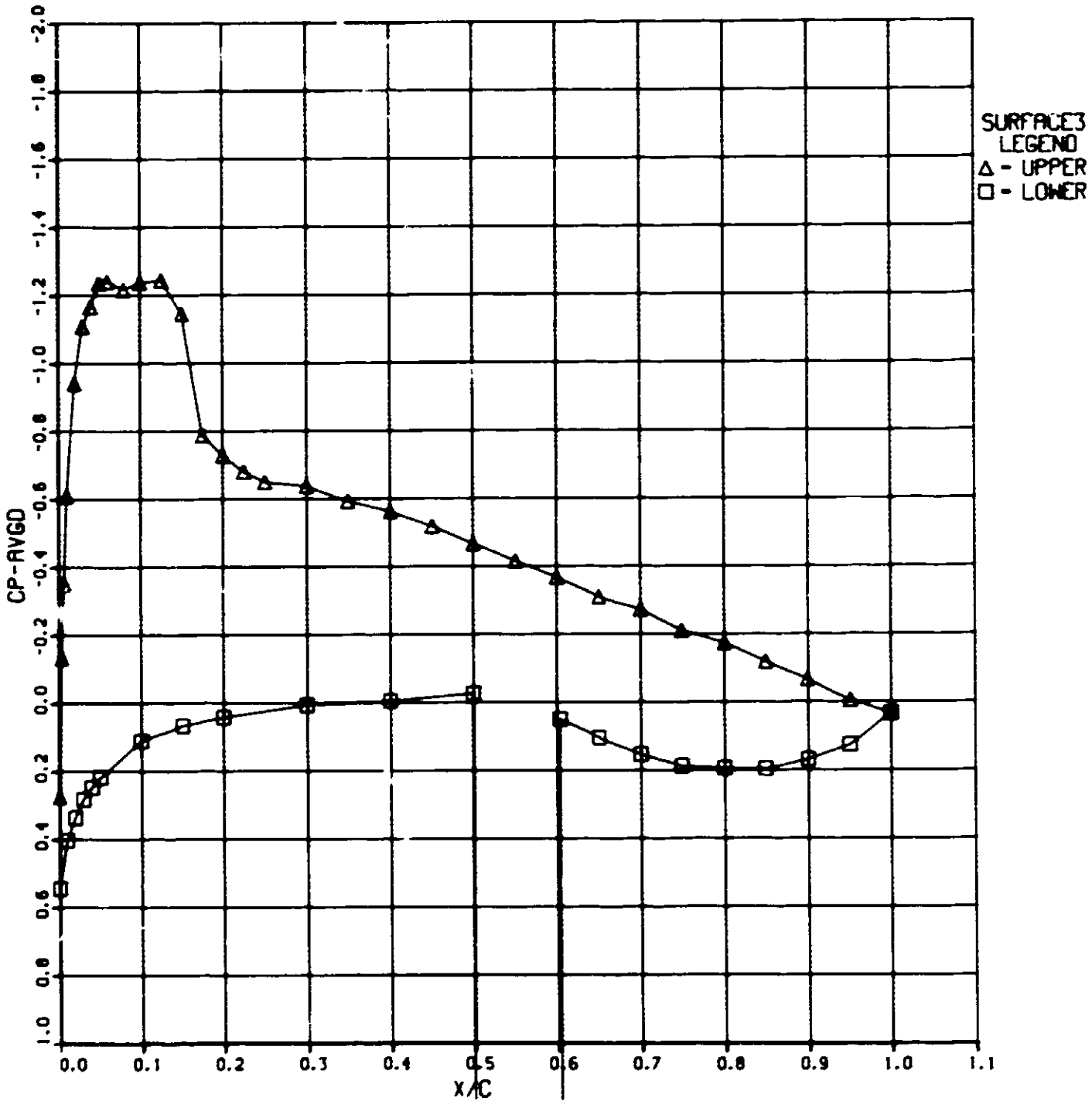

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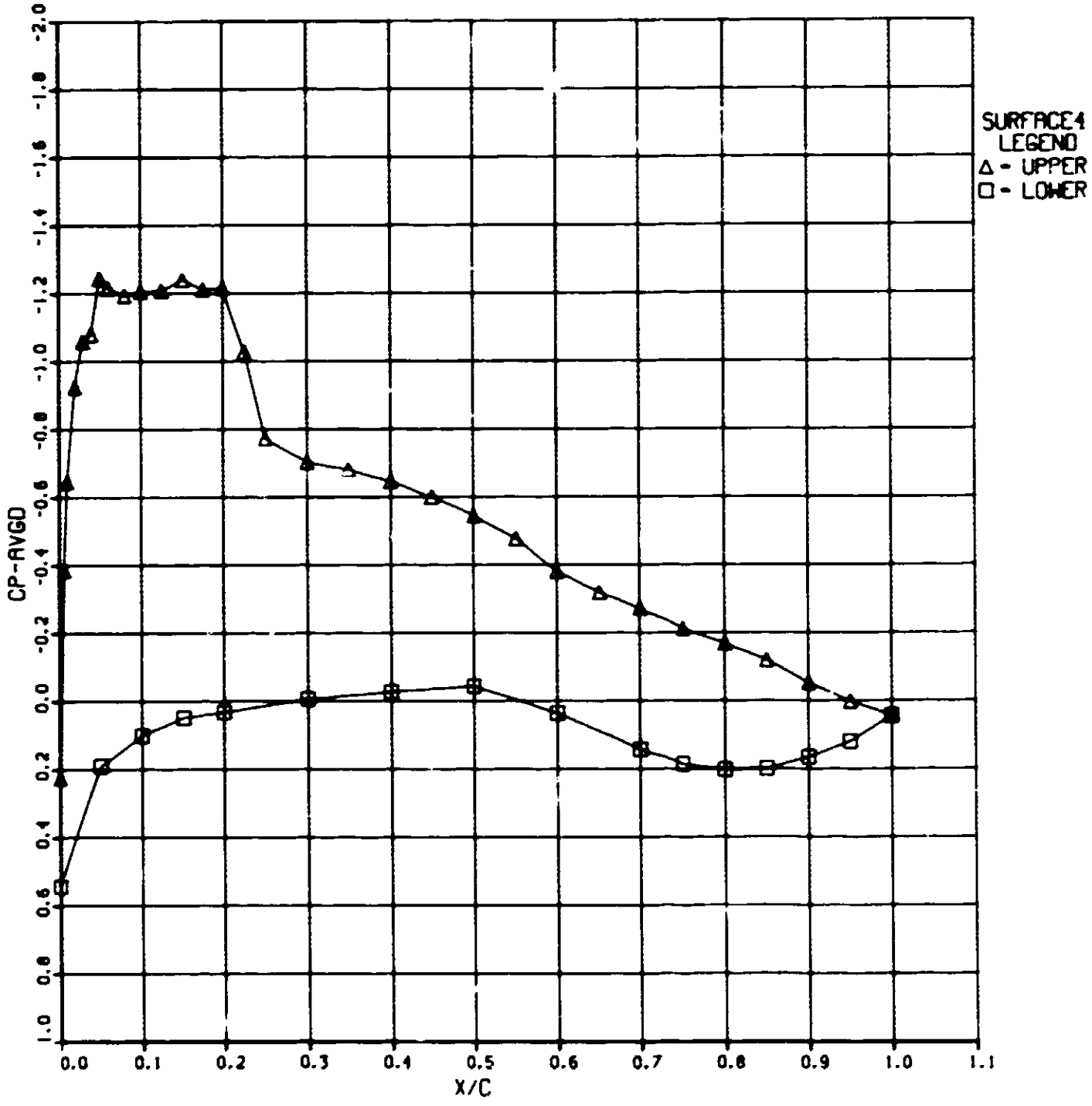

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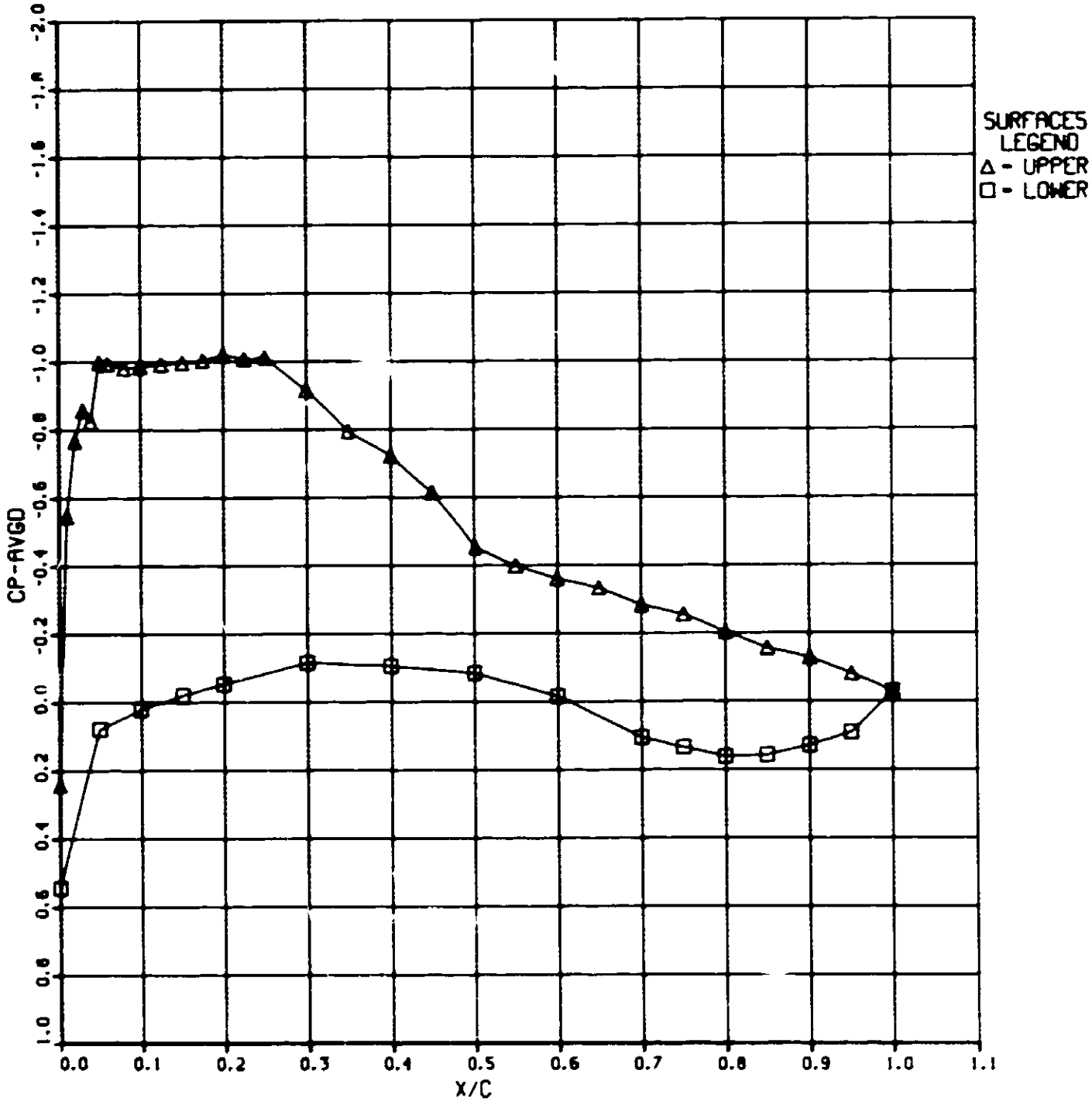

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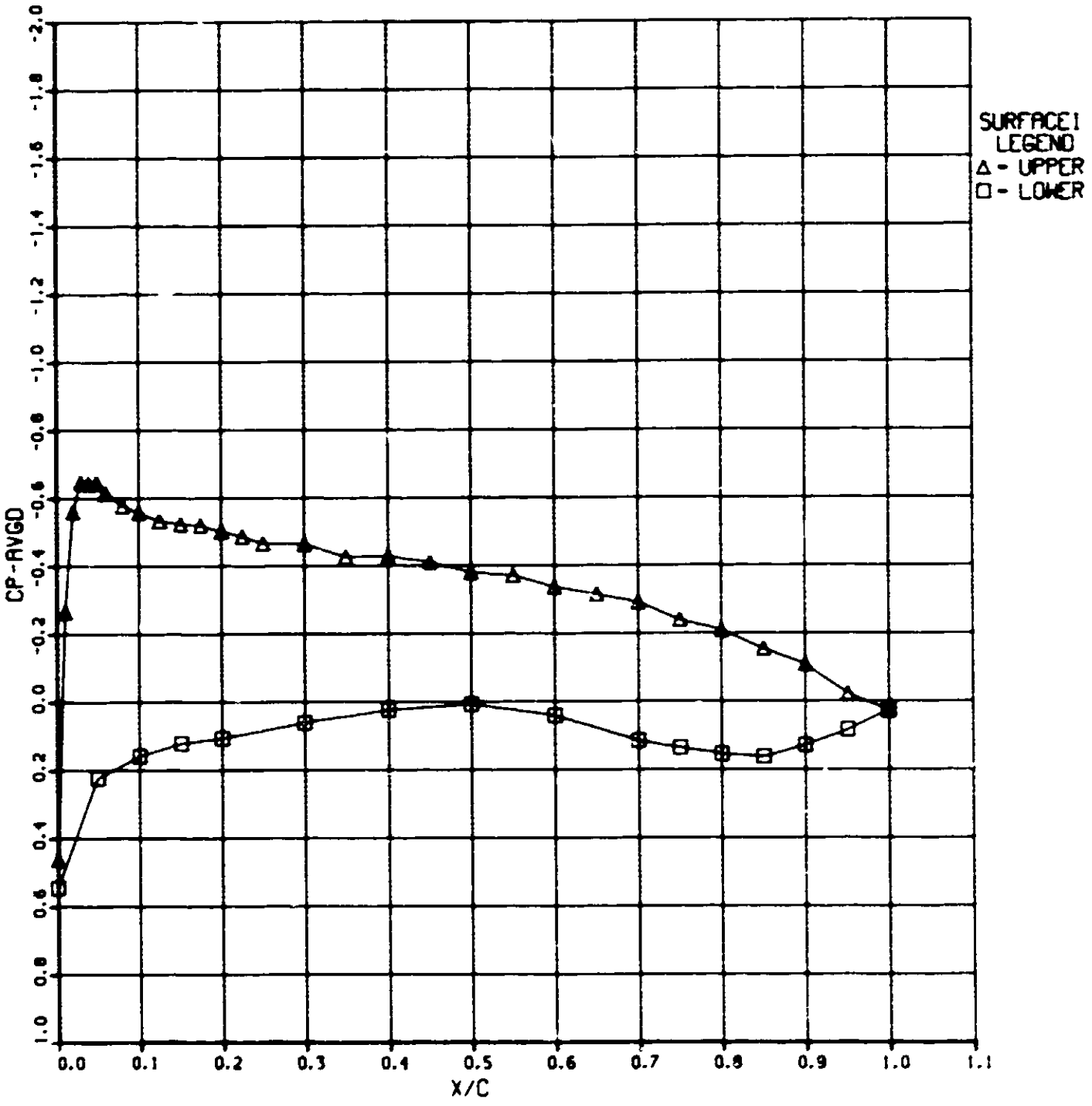

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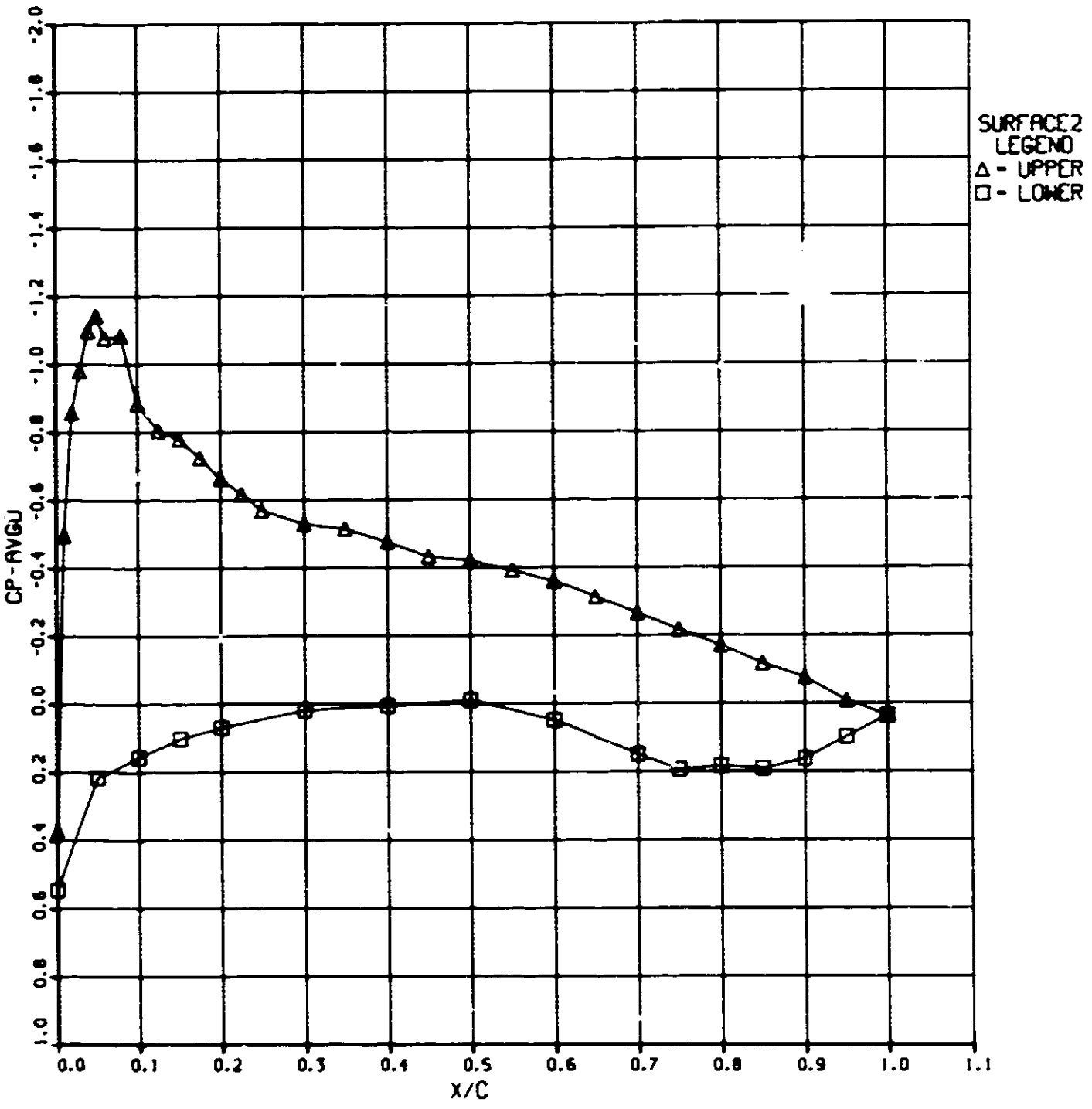

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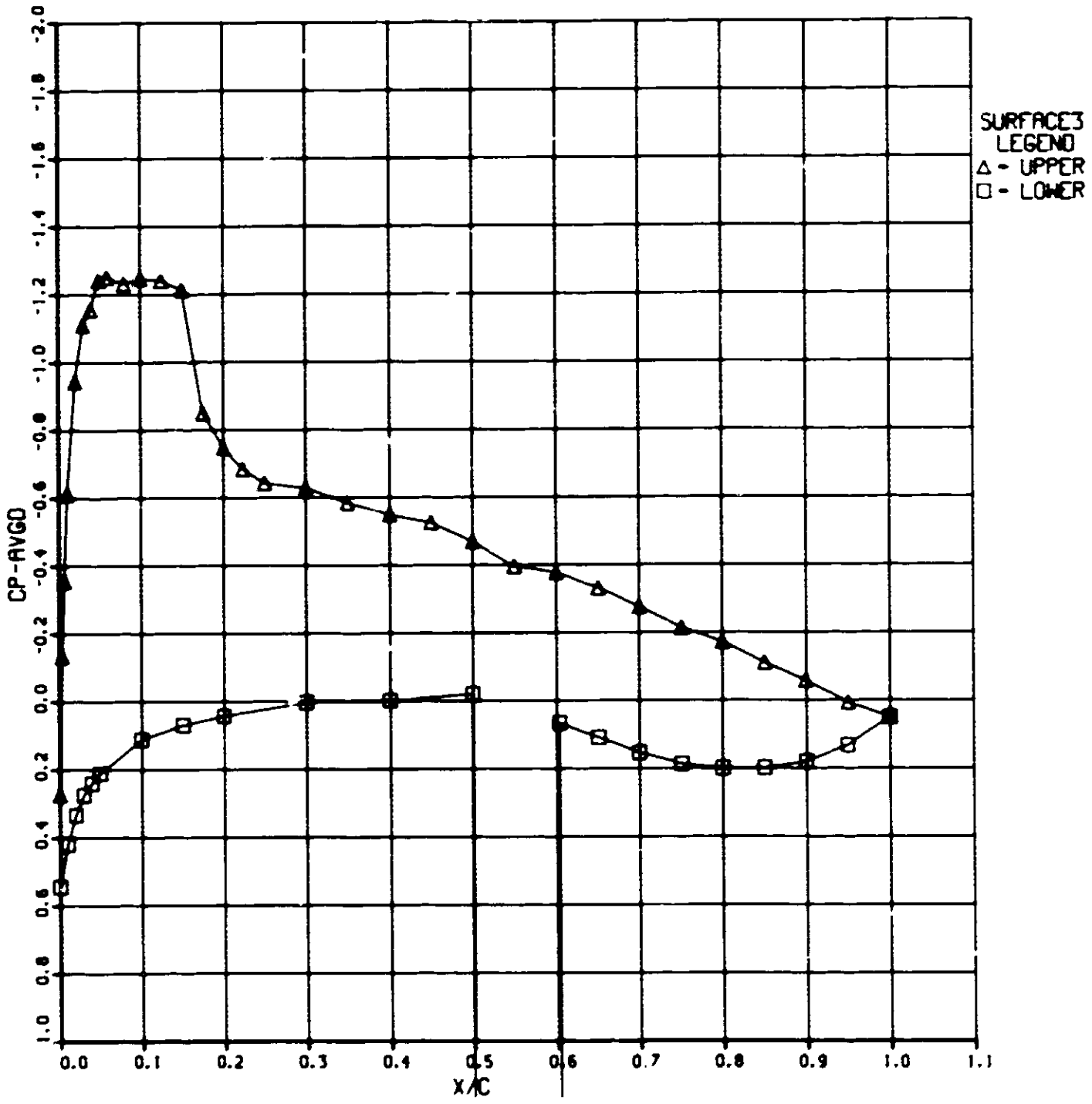

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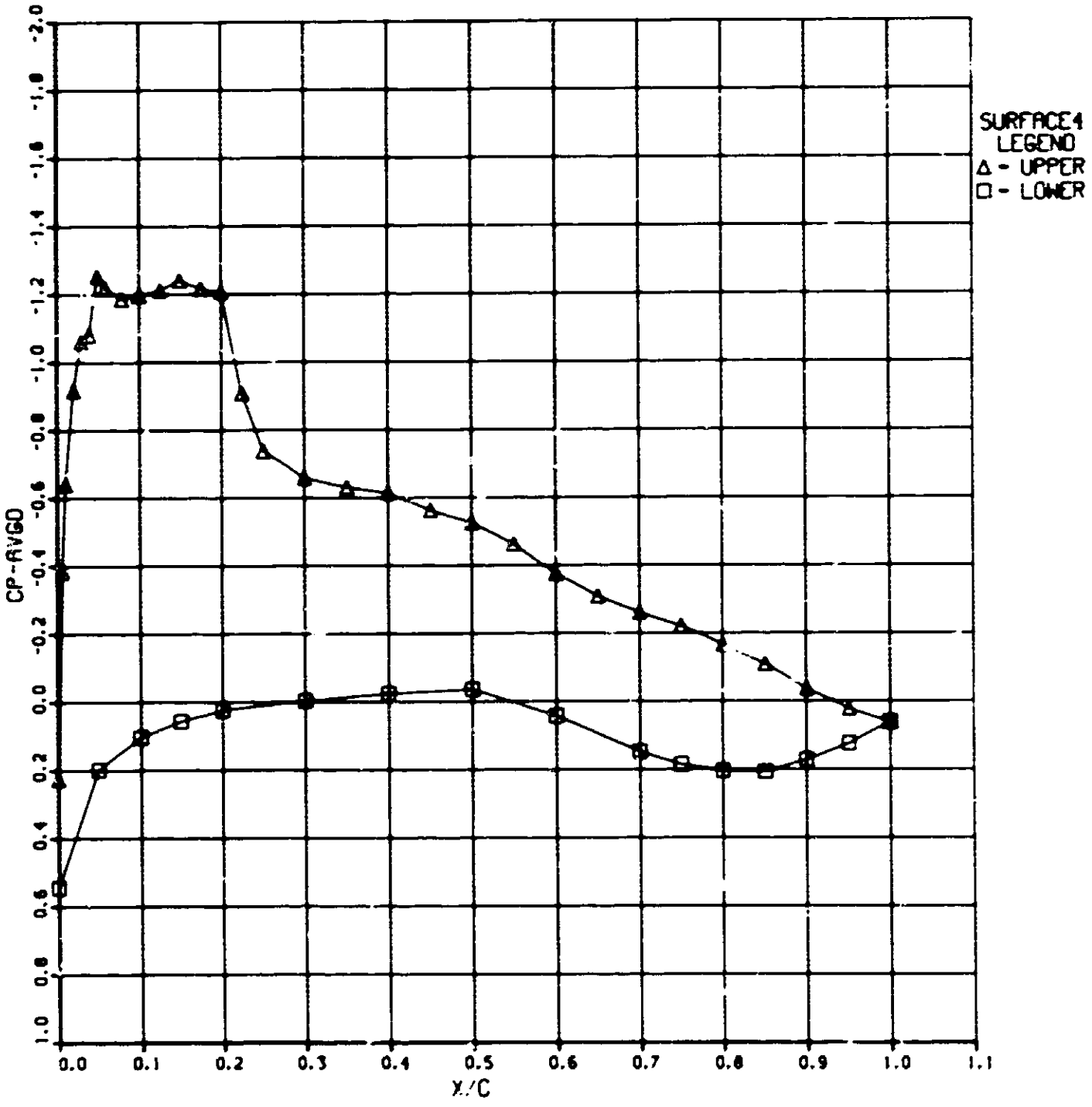

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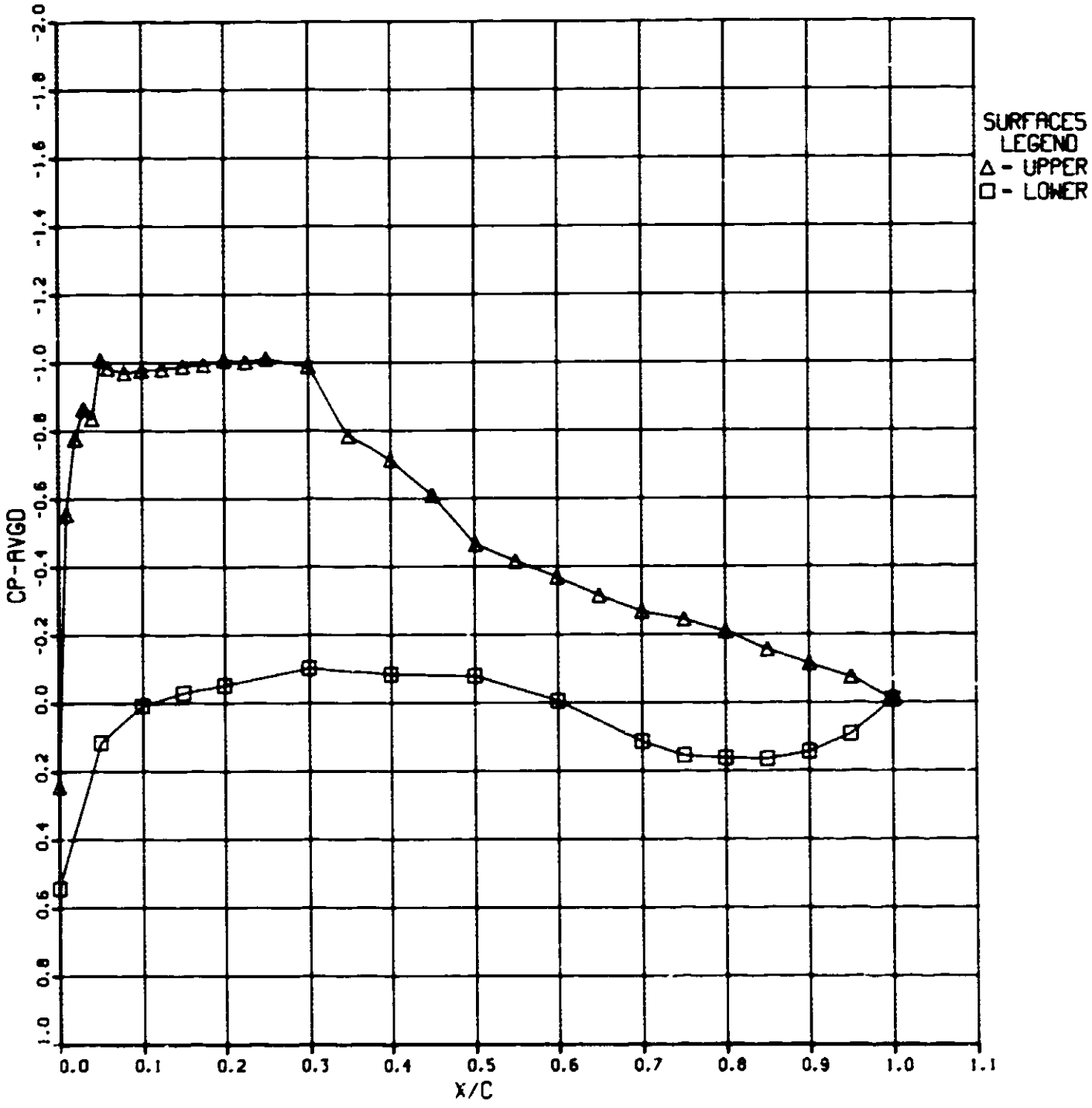
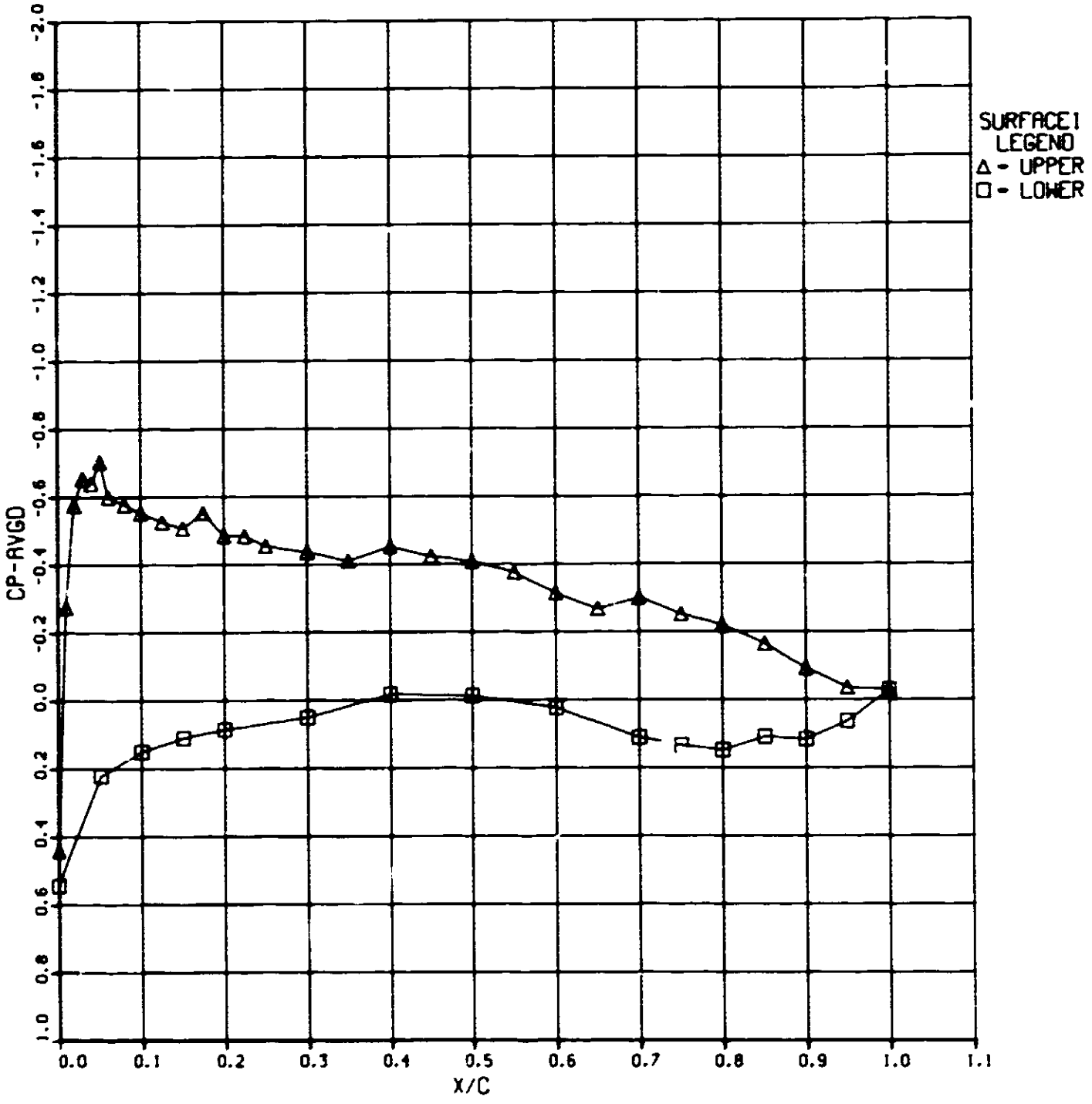


356-1-66 182.00: 2.00 CONF-17 MACH-0.818 RN-1.994 PT-1018 ALPHA- 5.00



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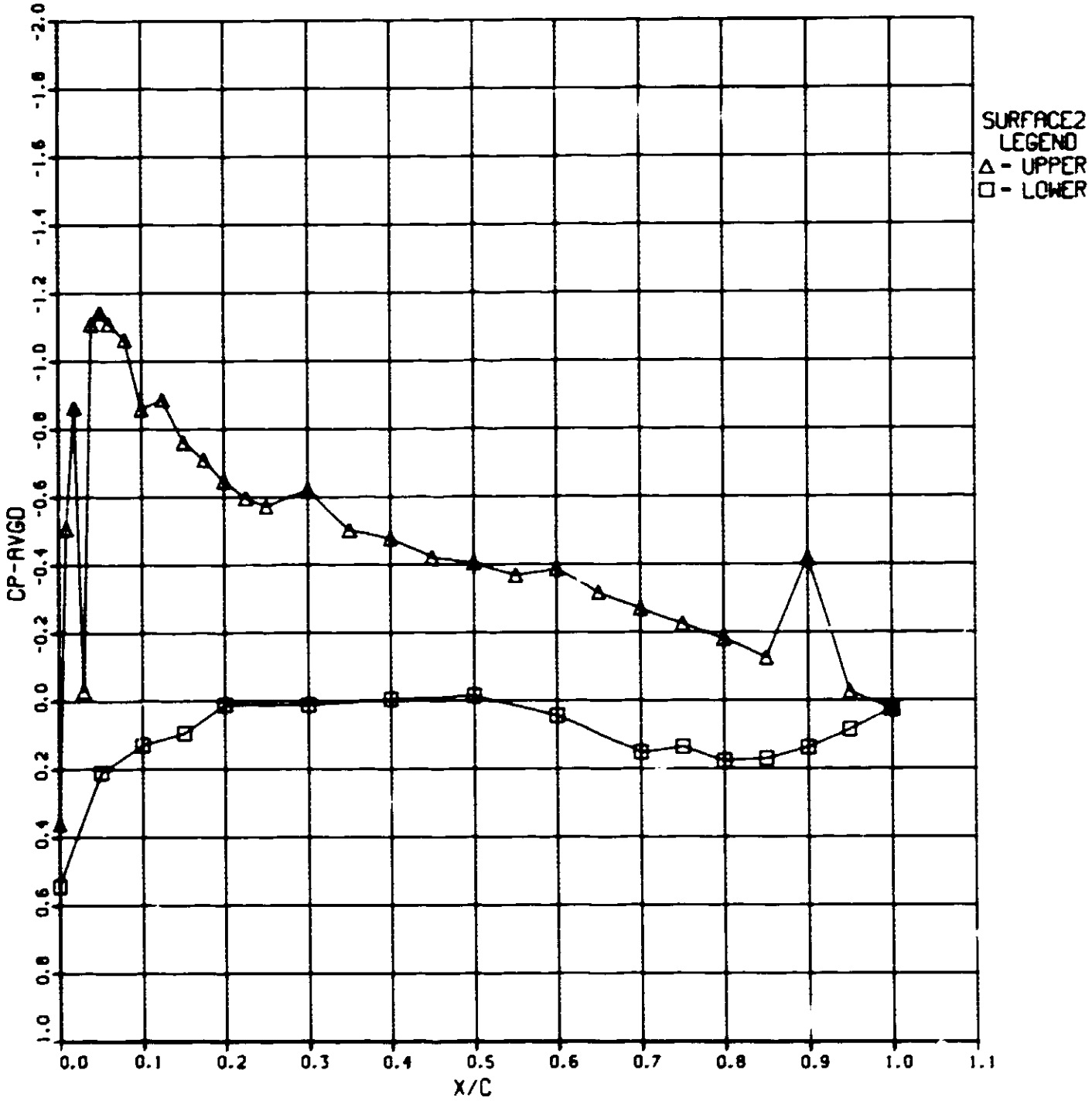

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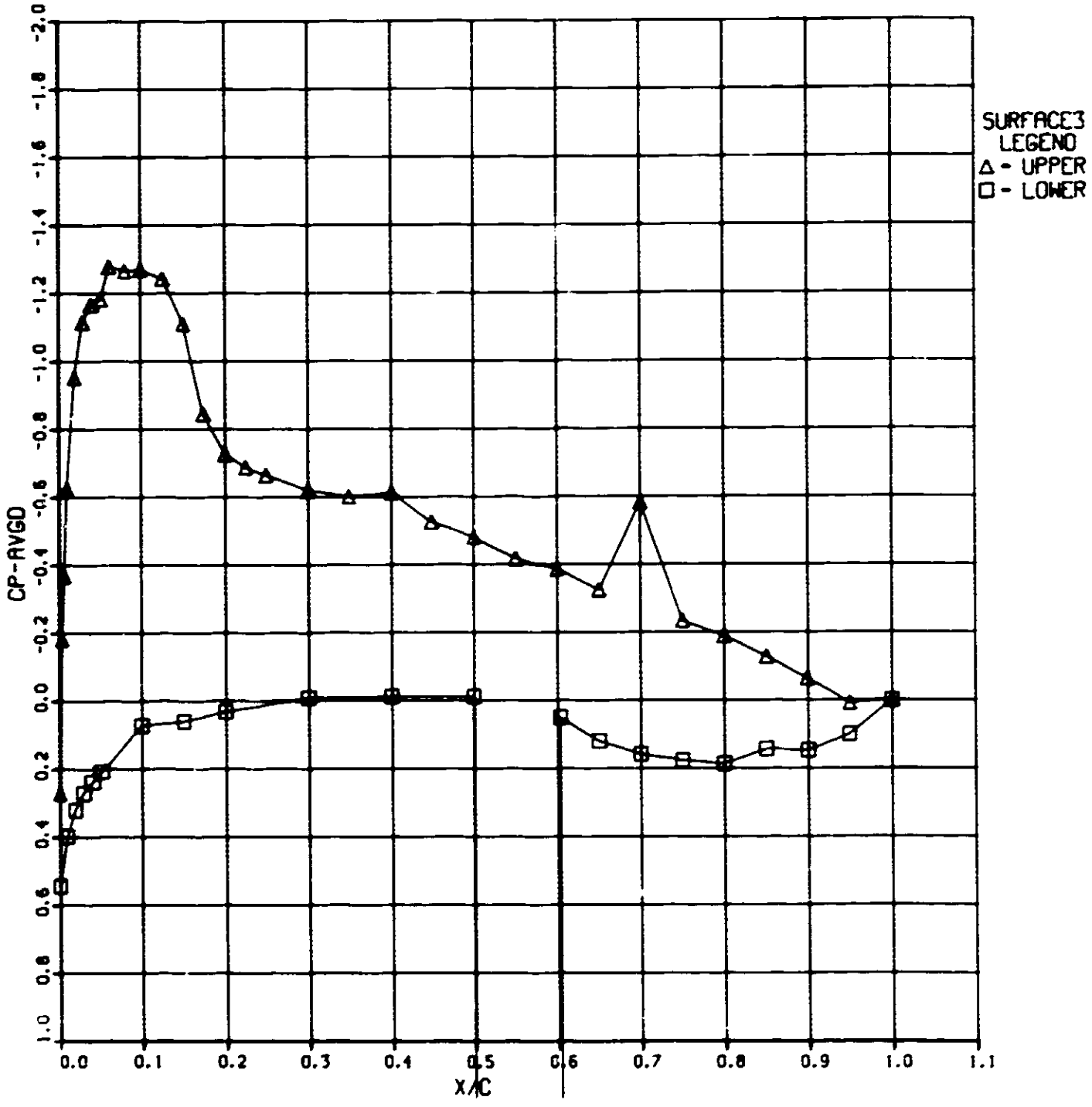

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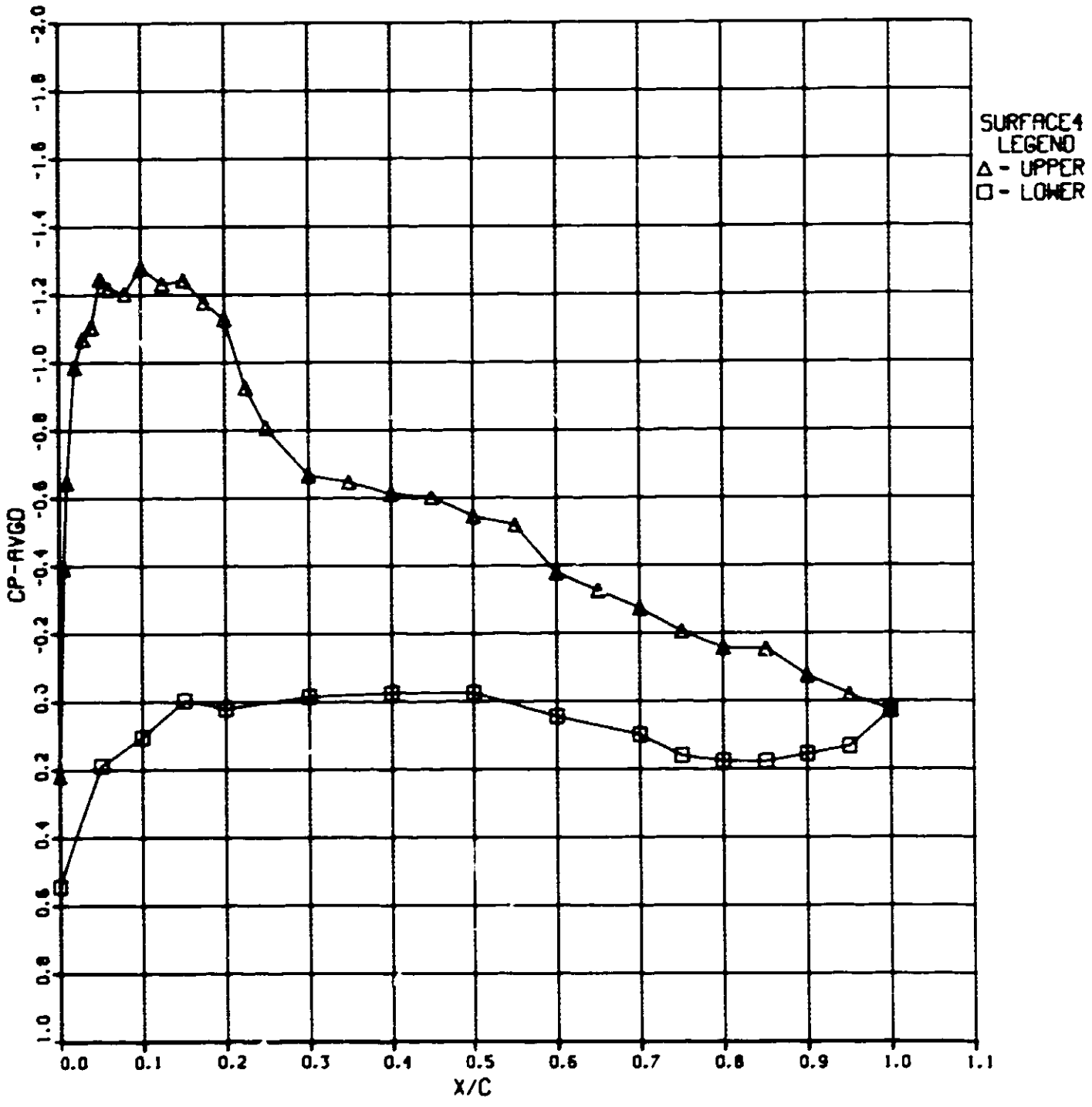

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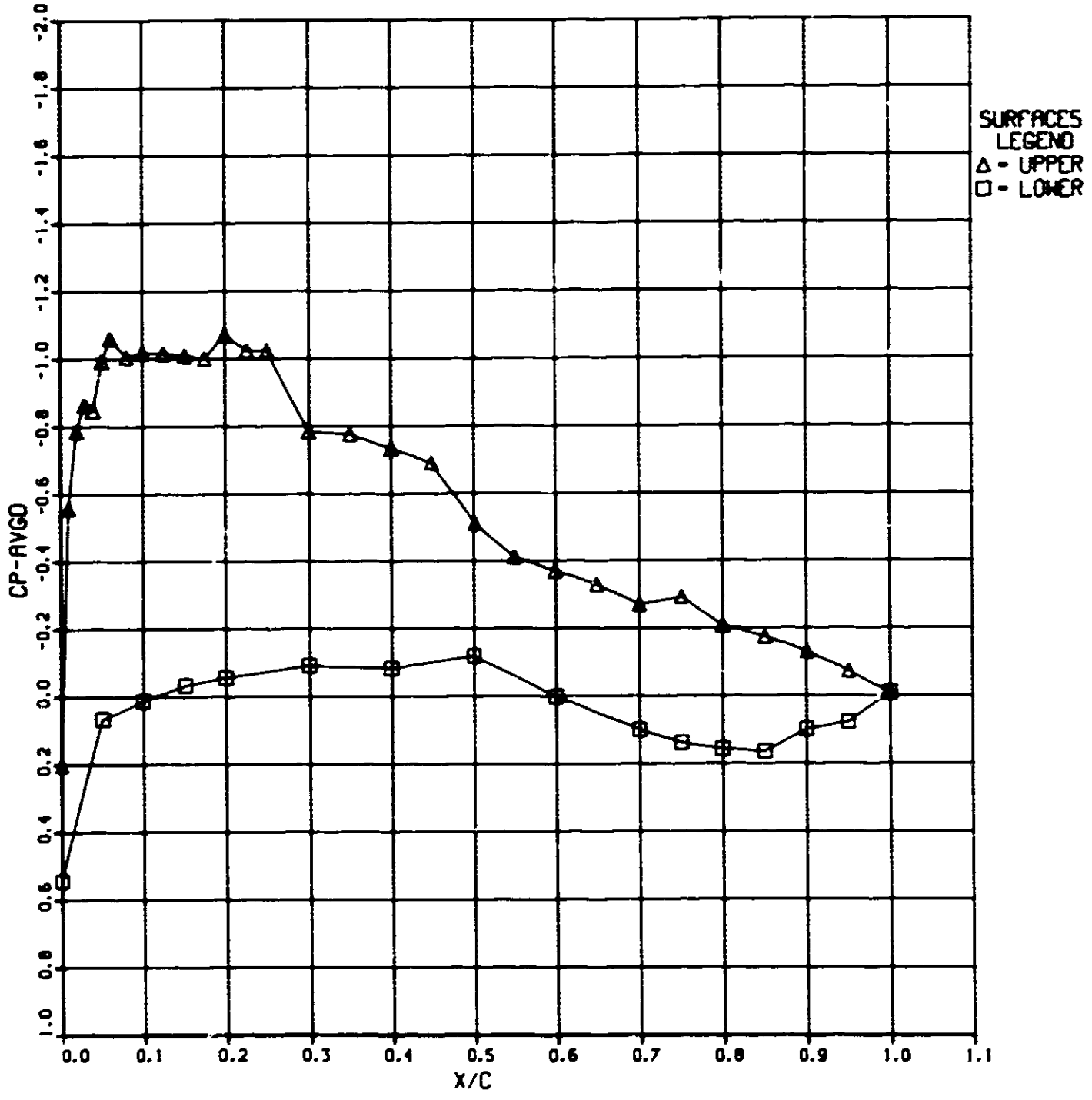

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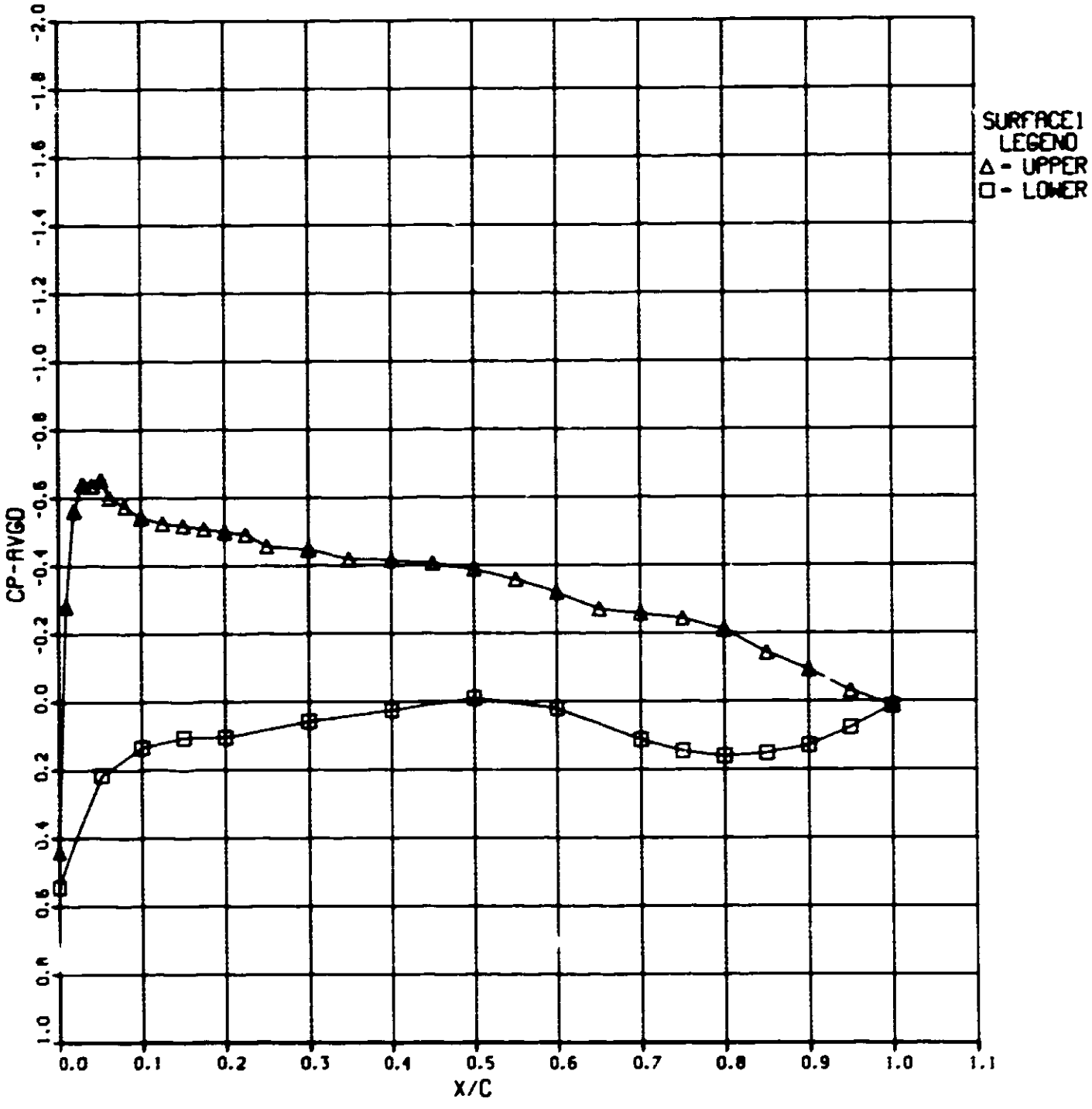

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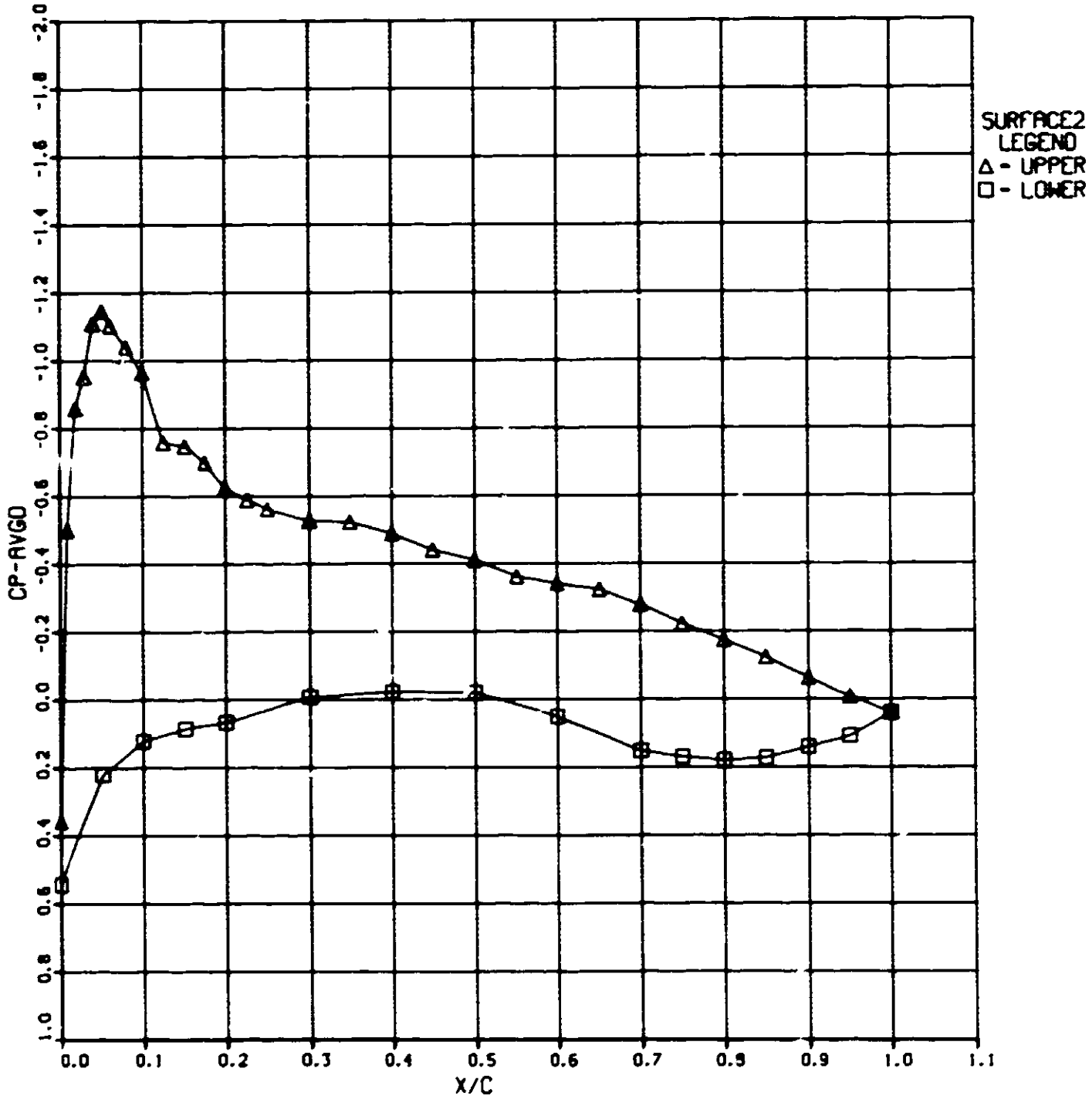

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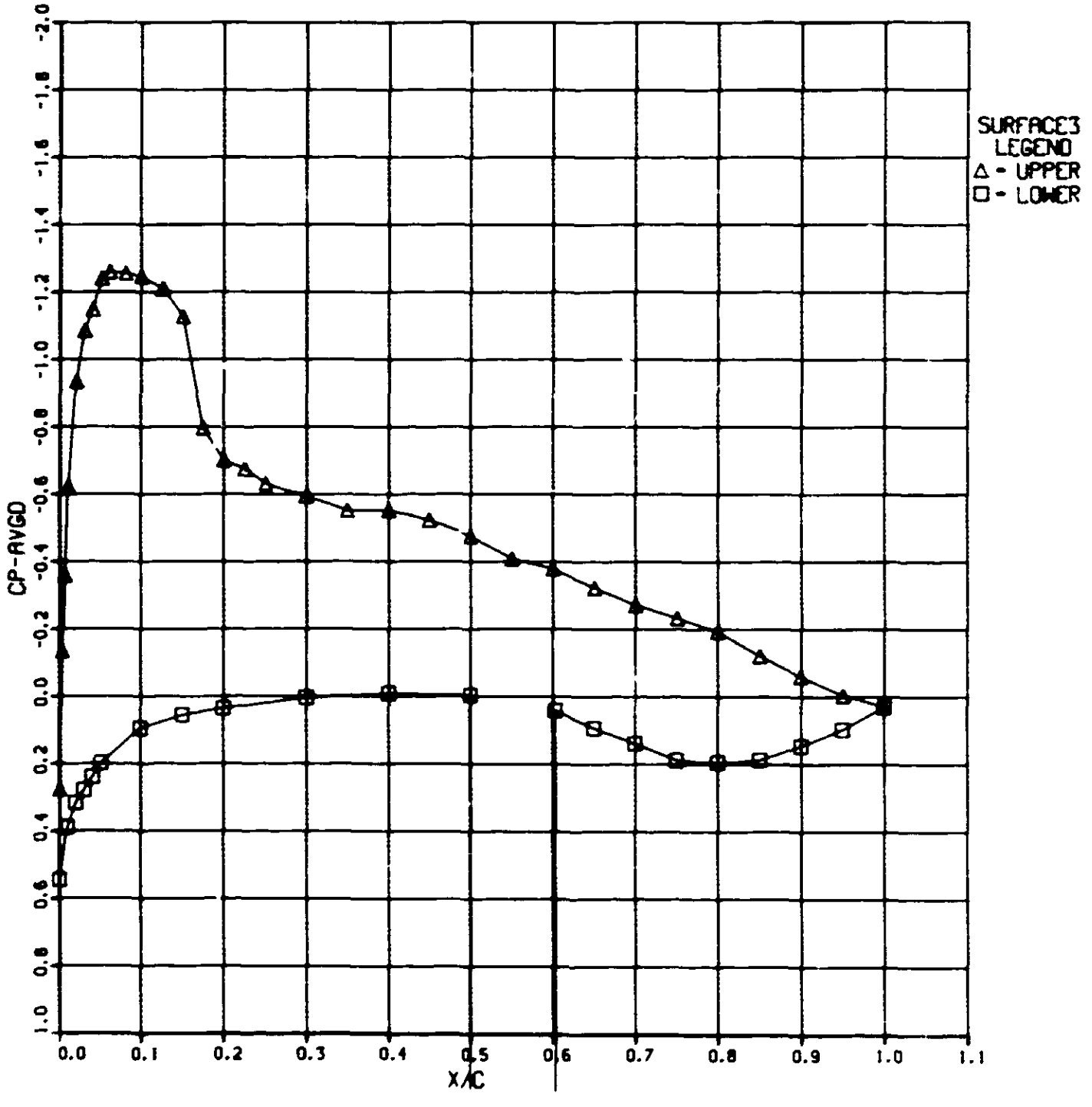

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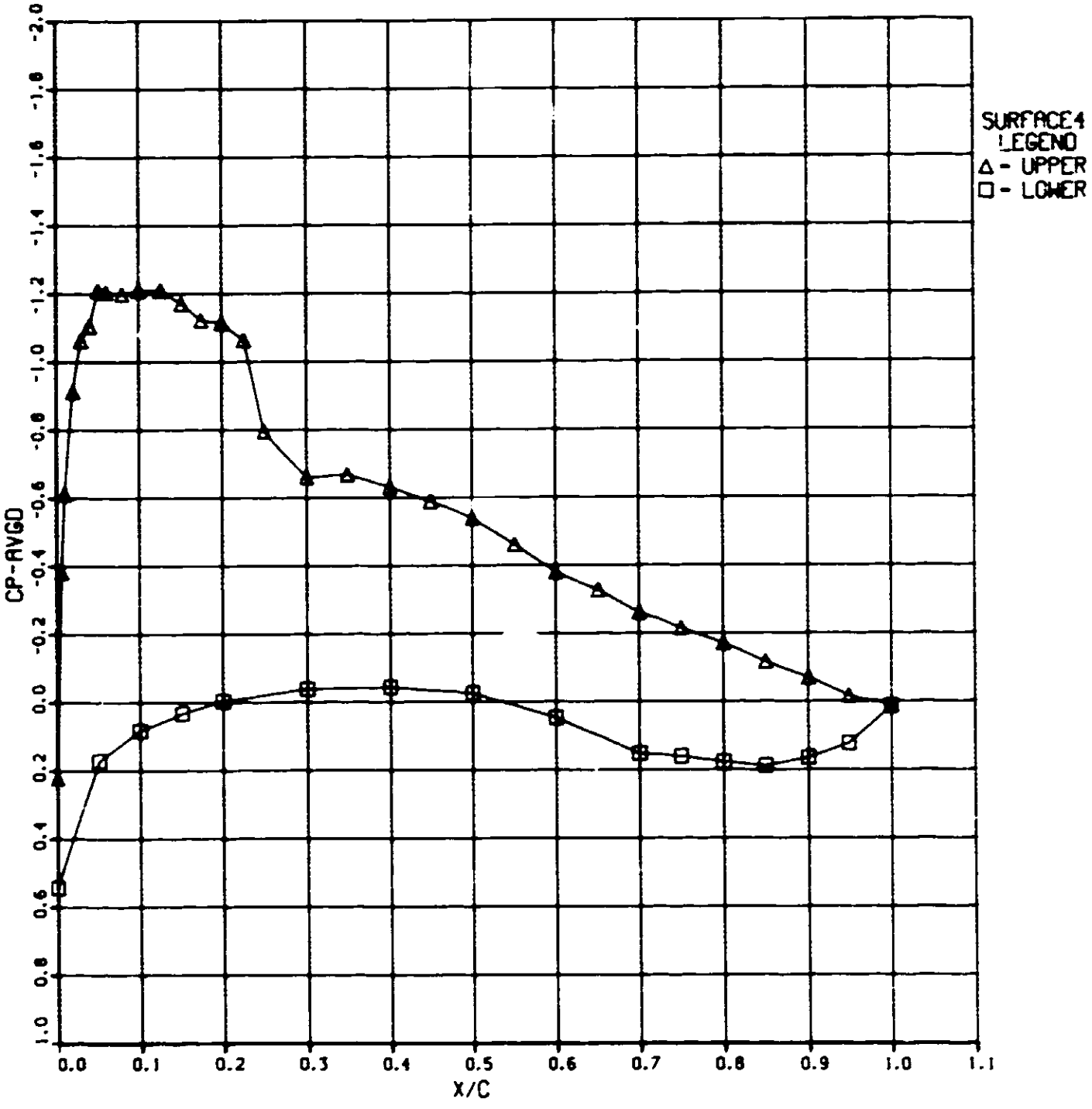

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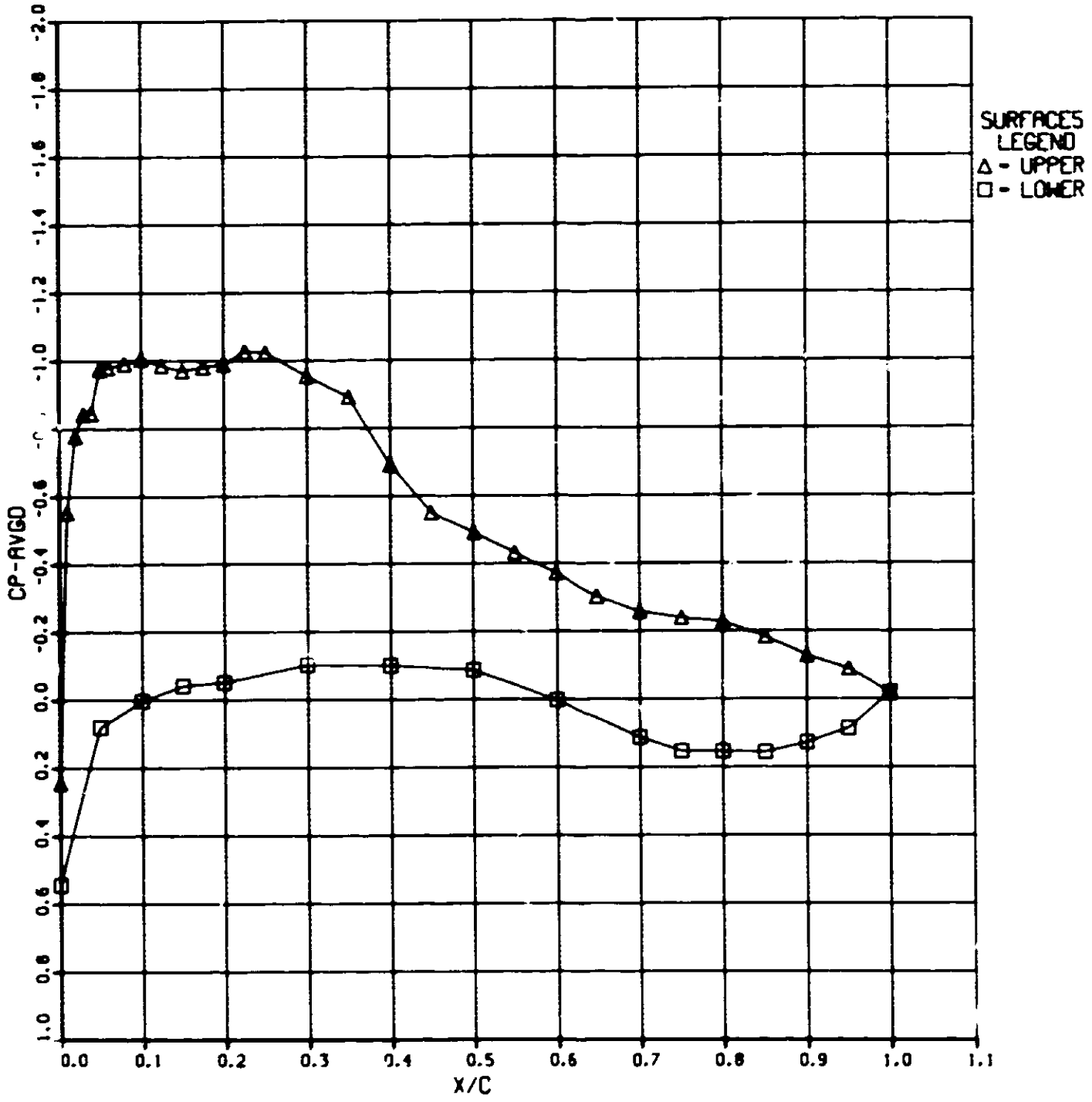

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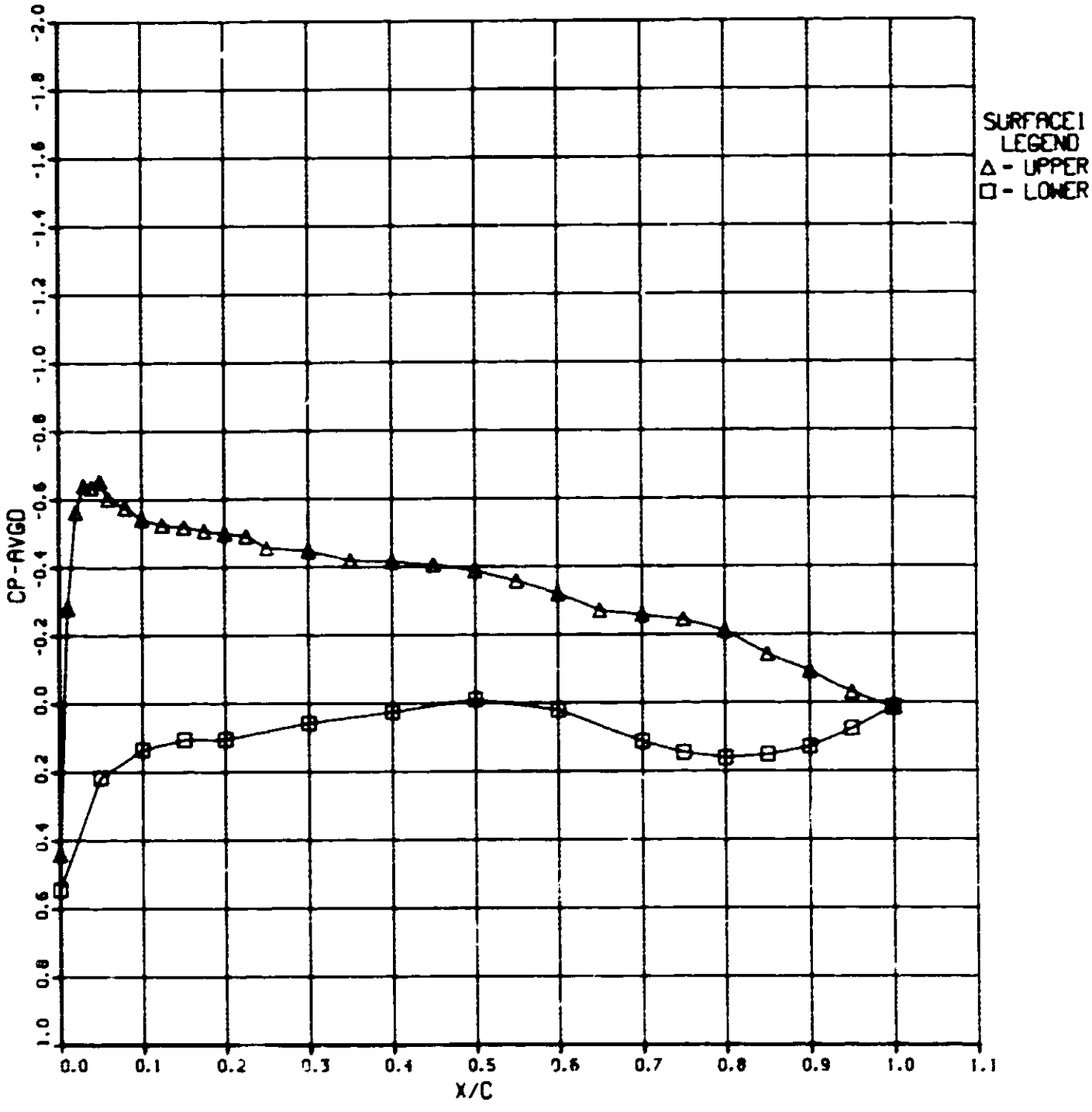

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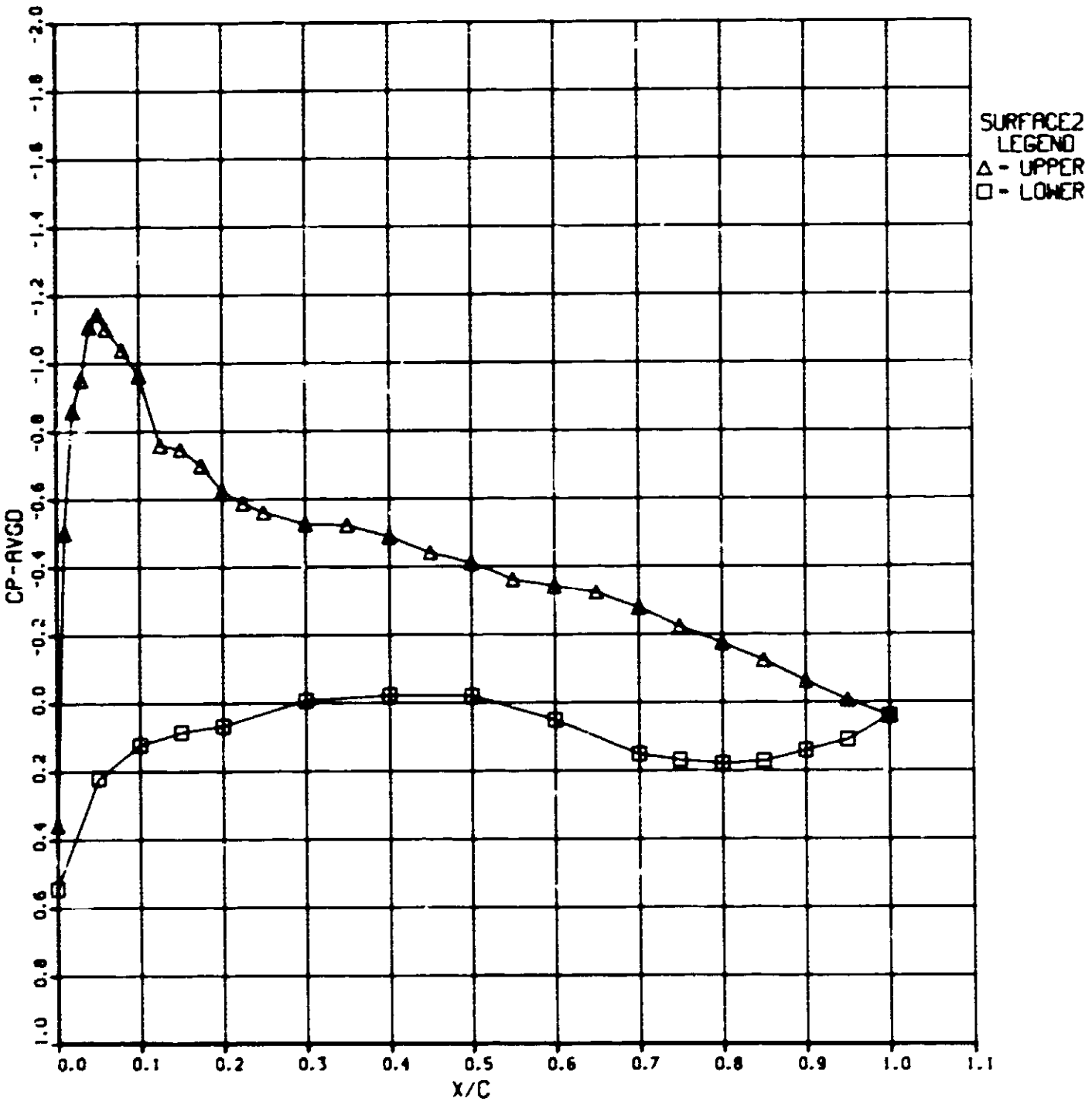

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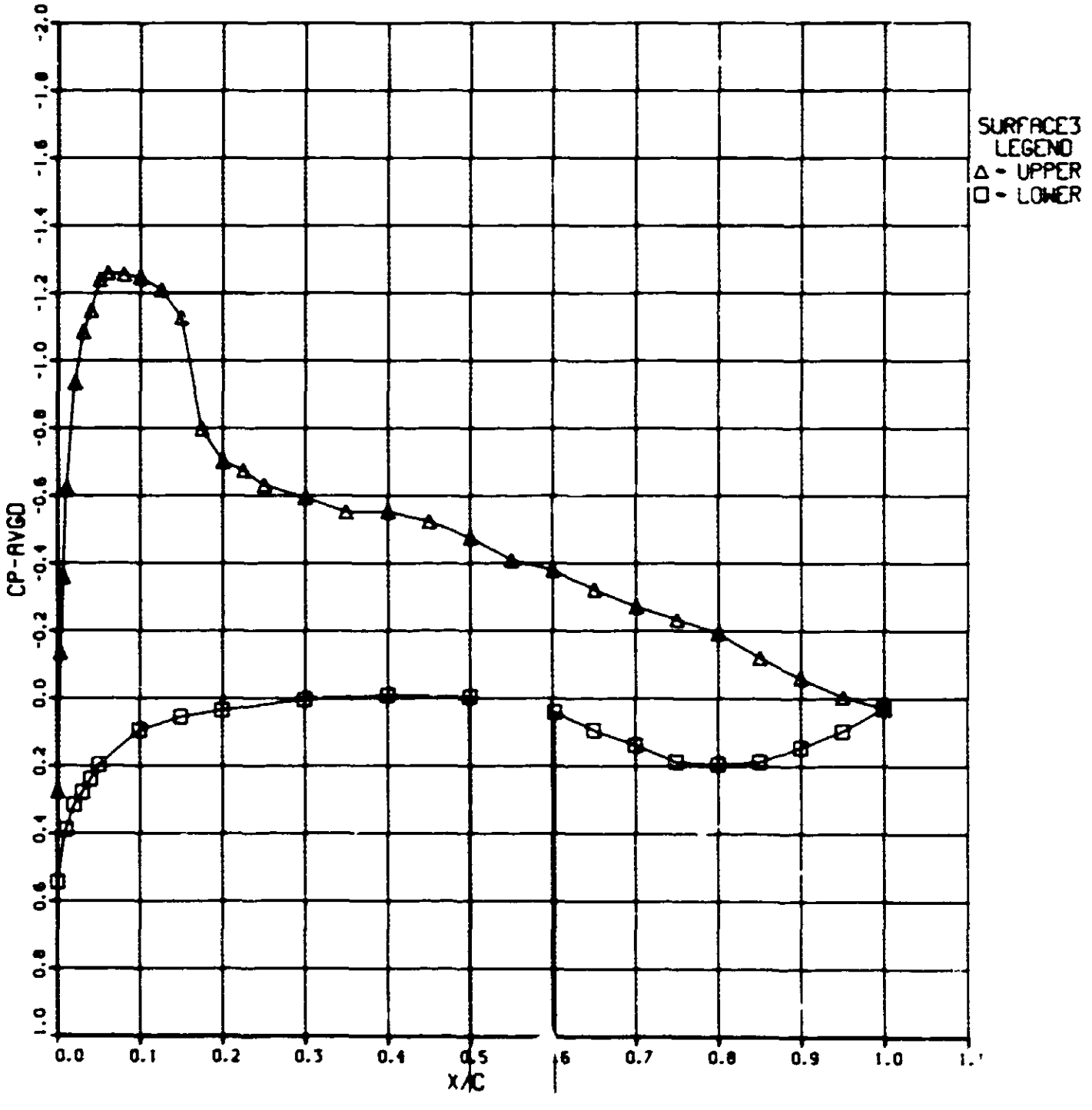

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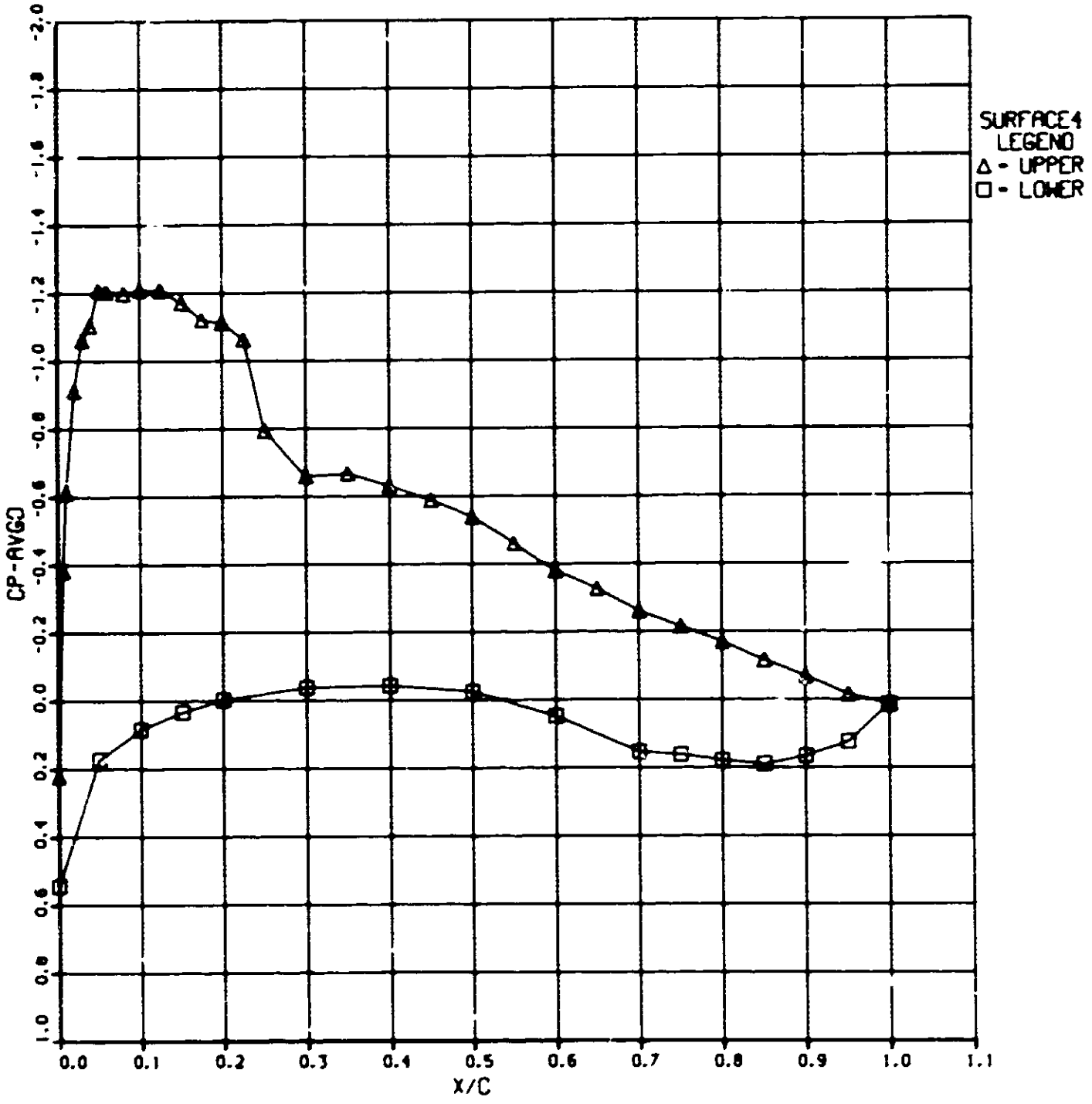

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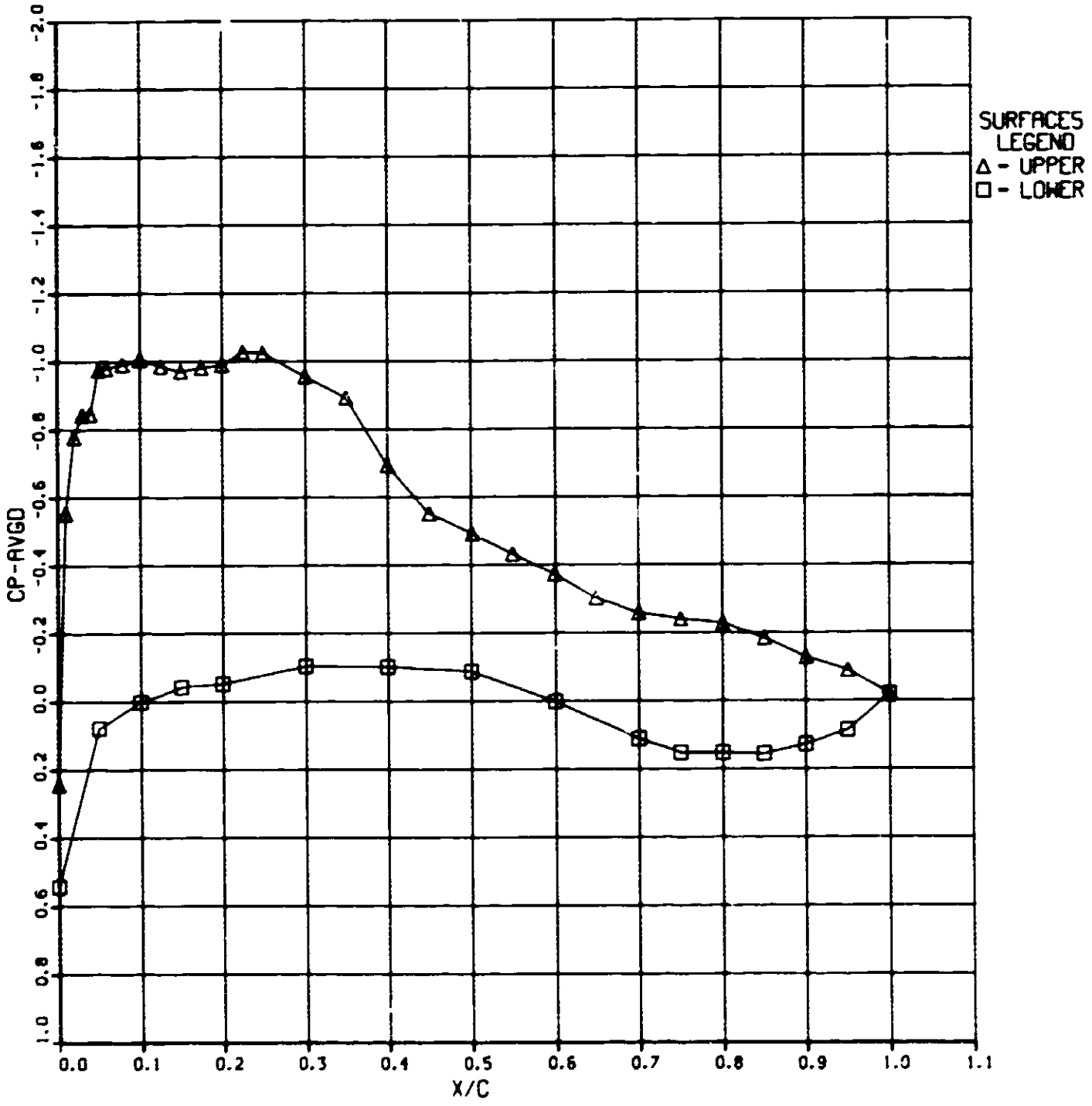

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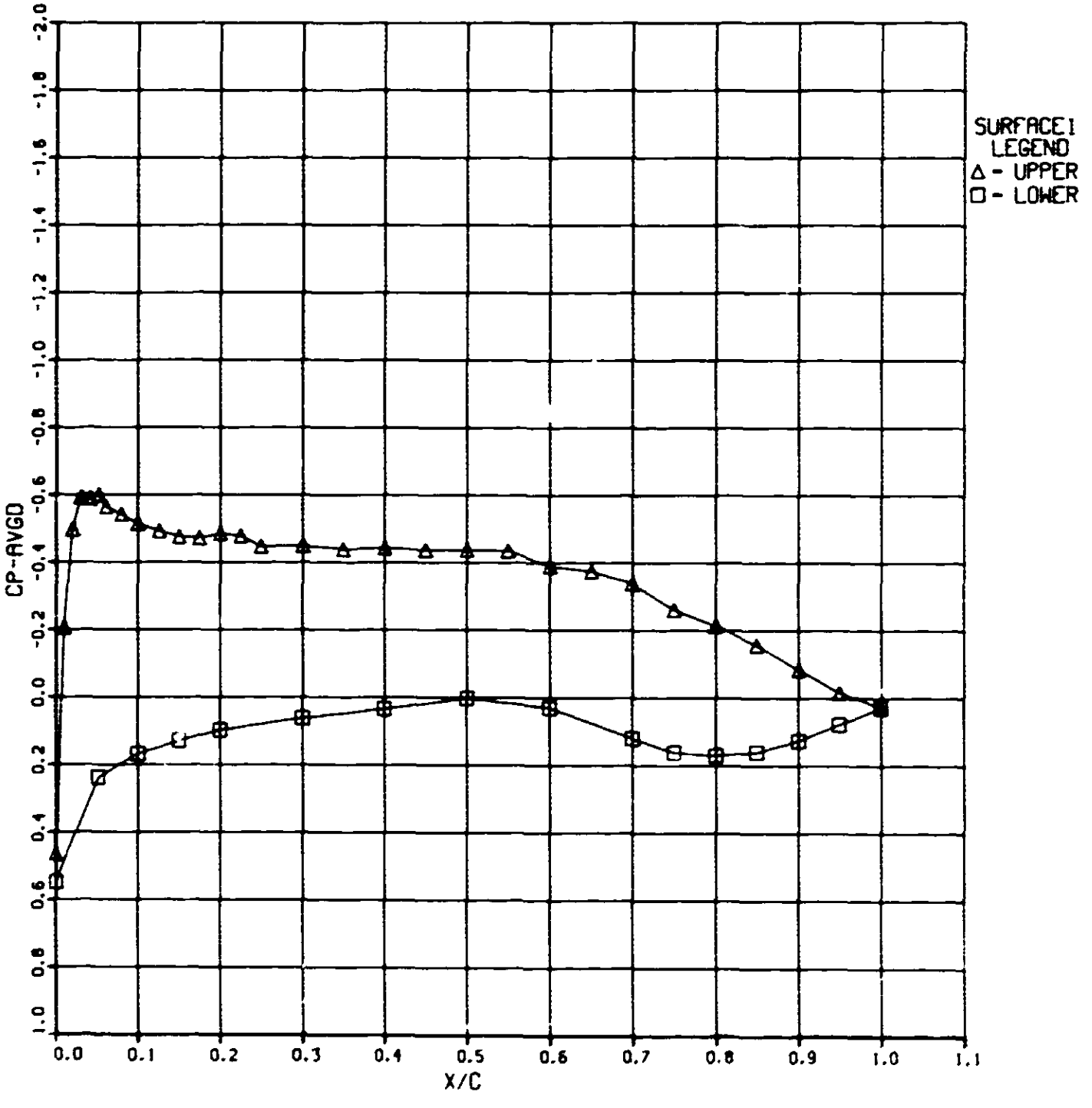

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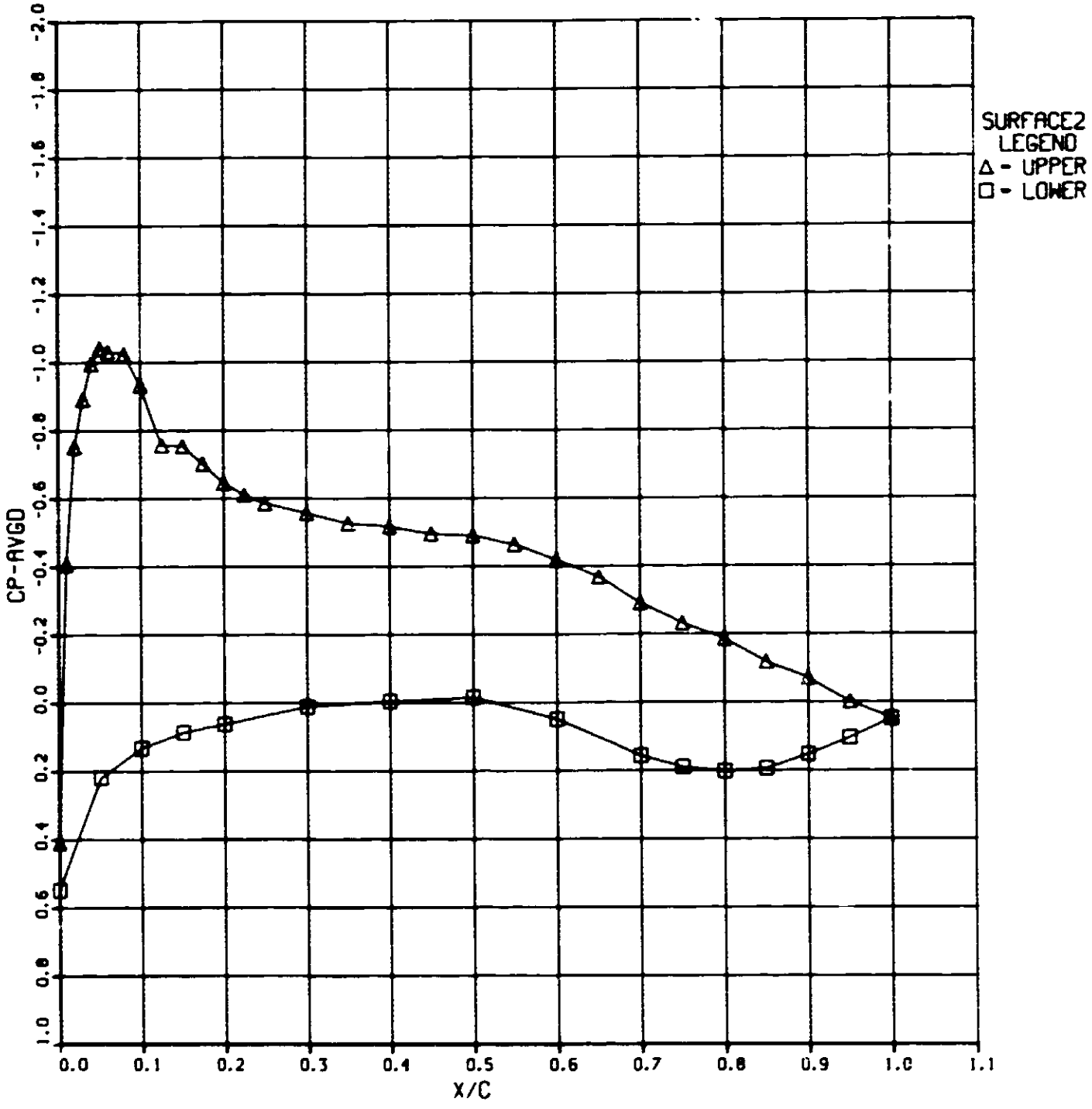

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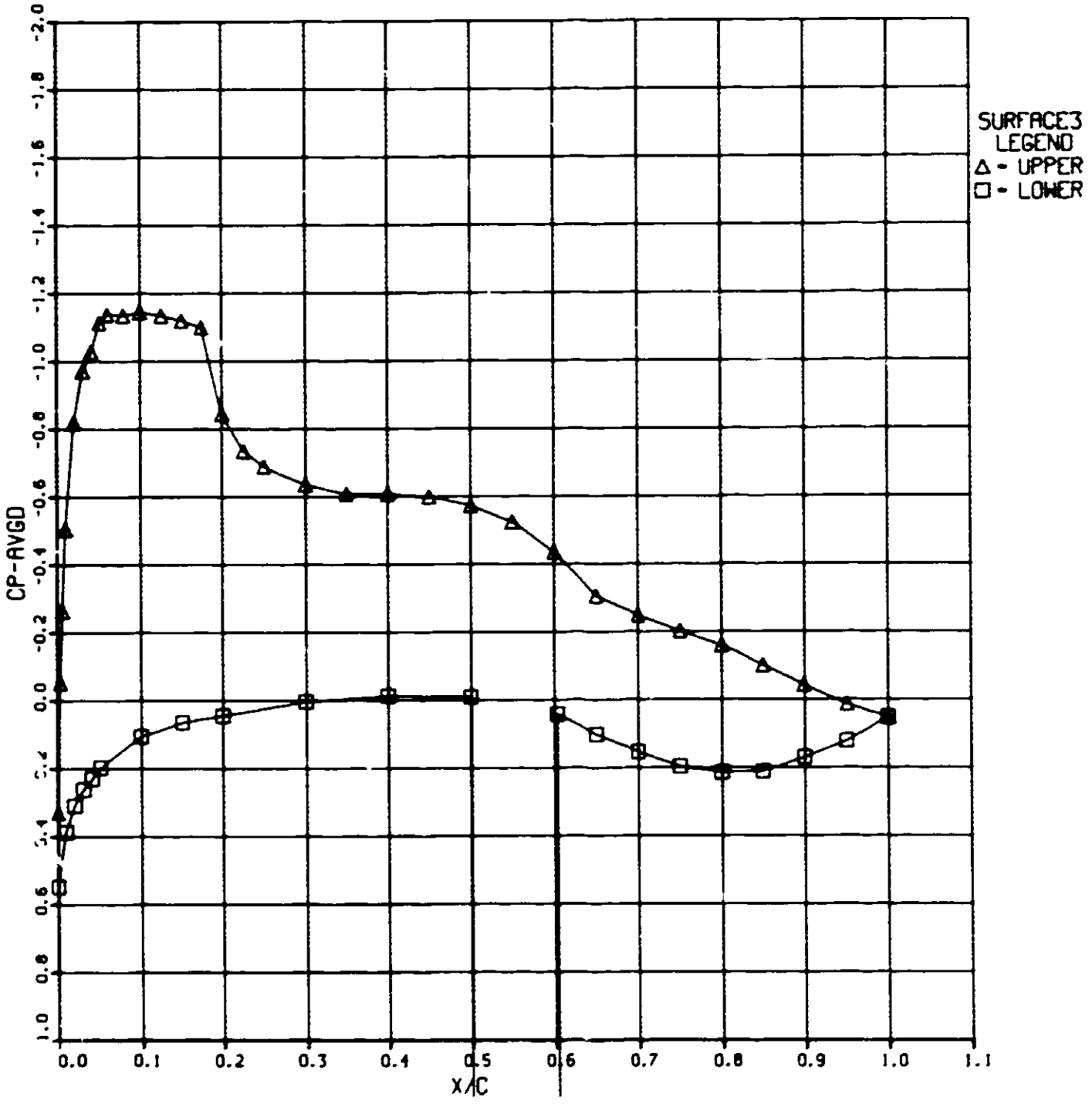

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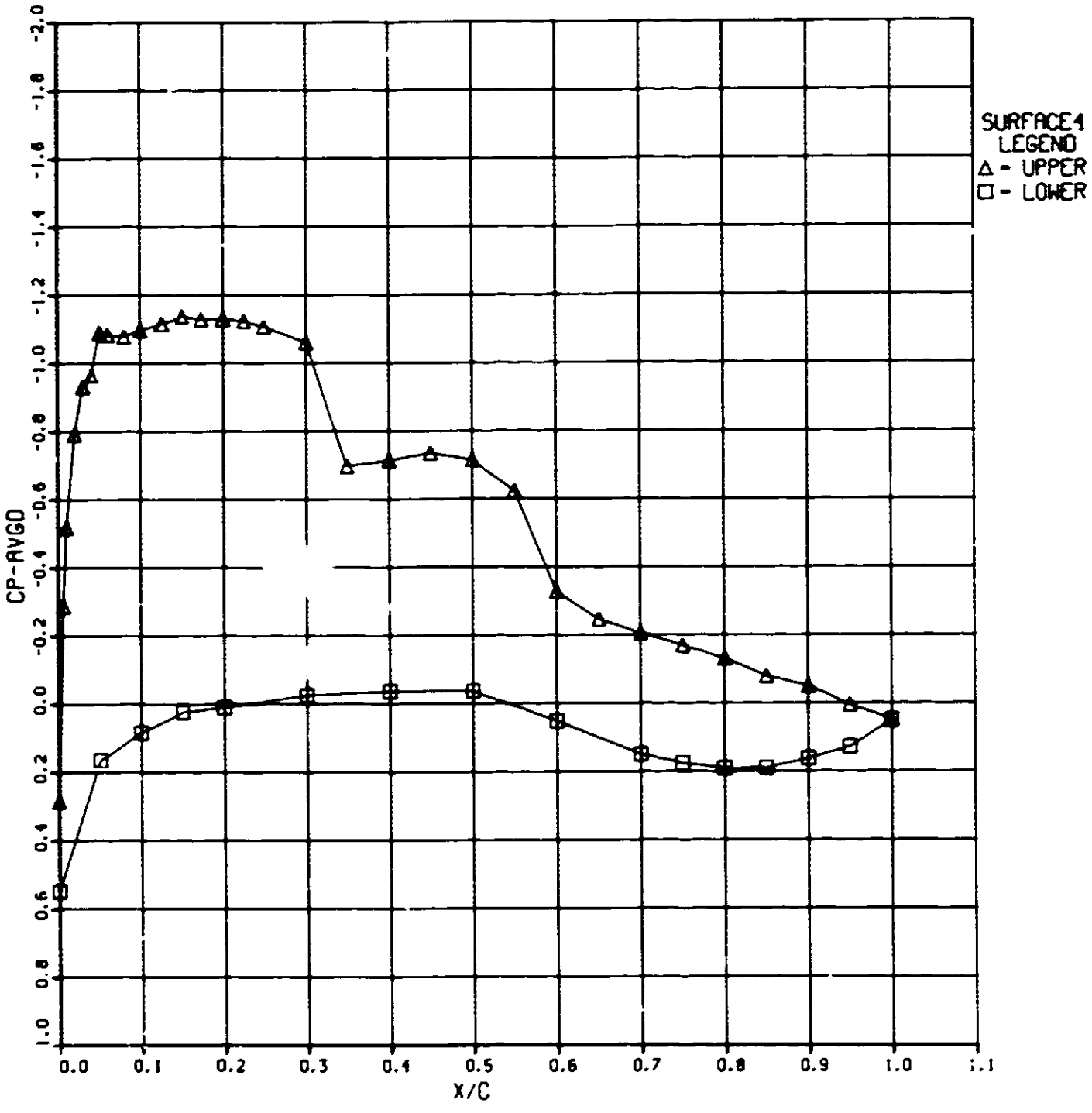

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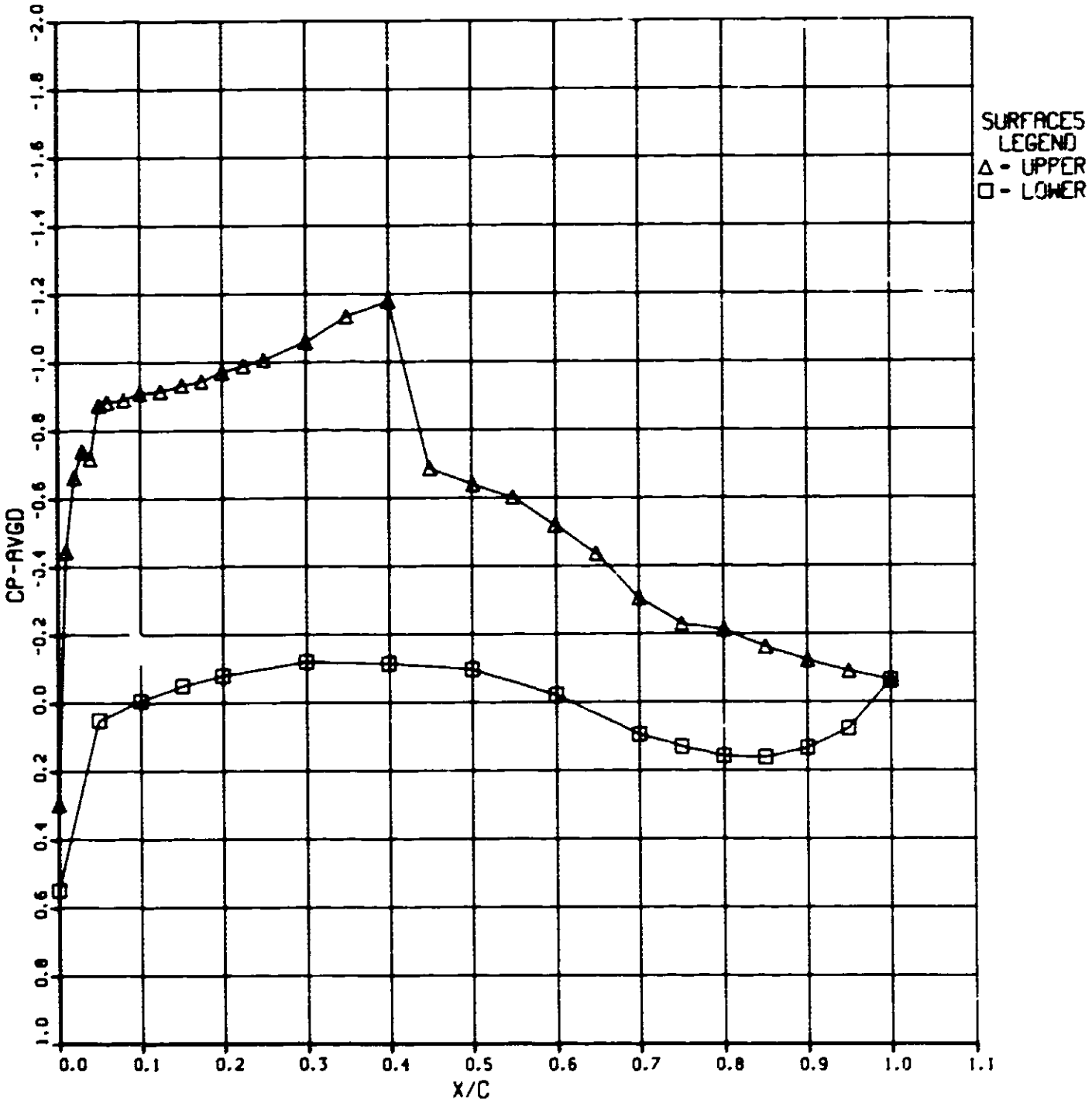

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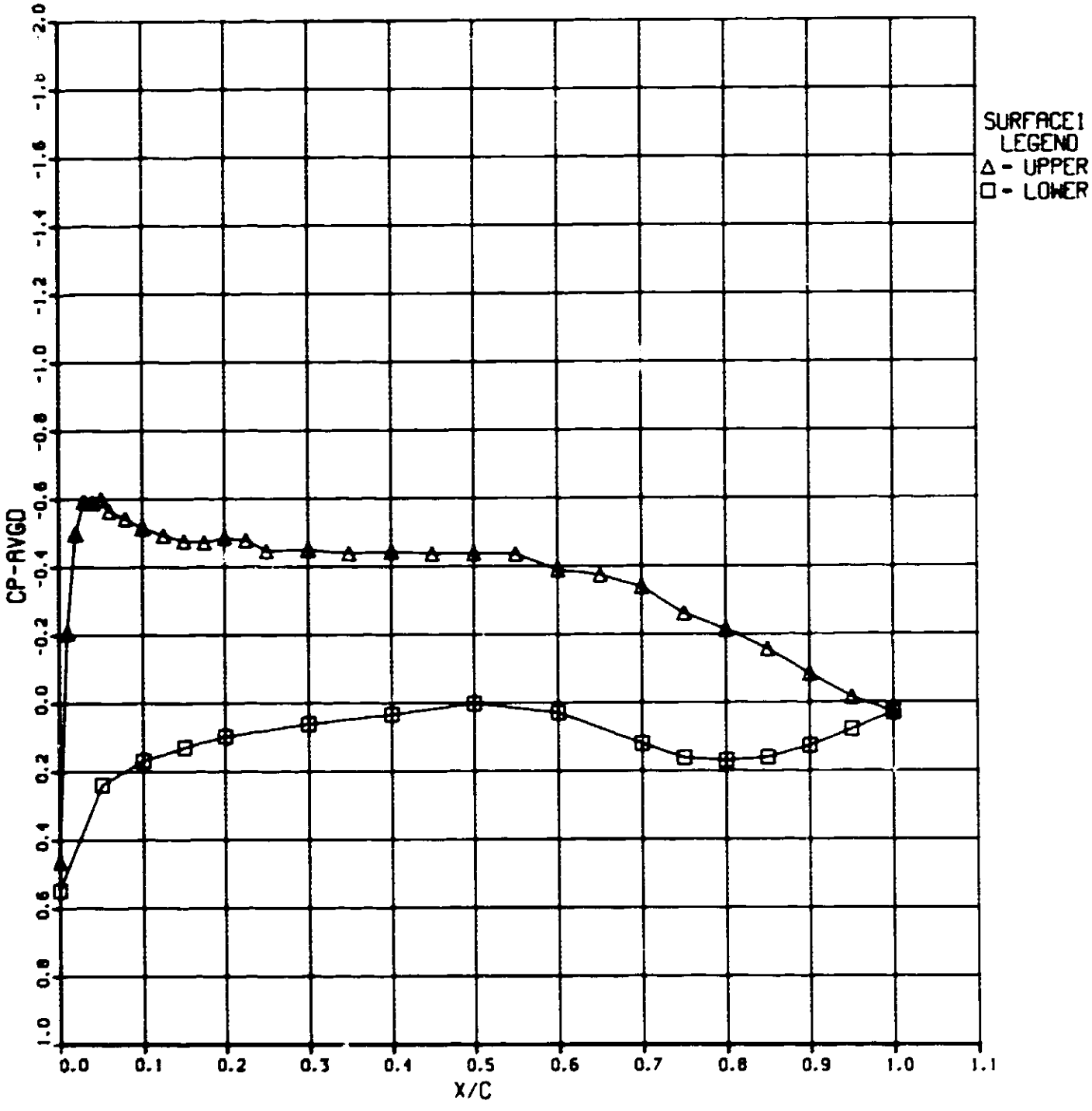

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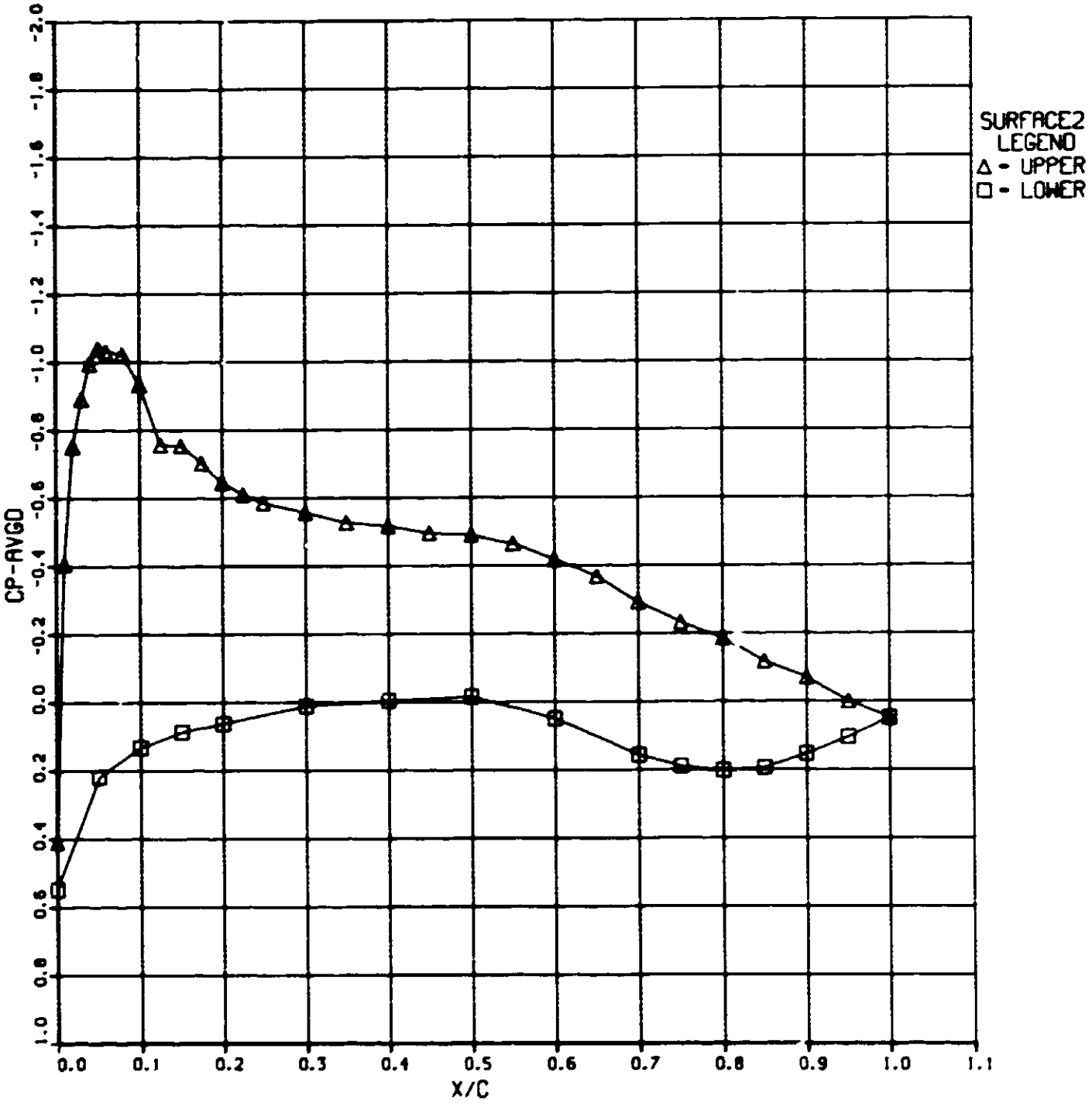

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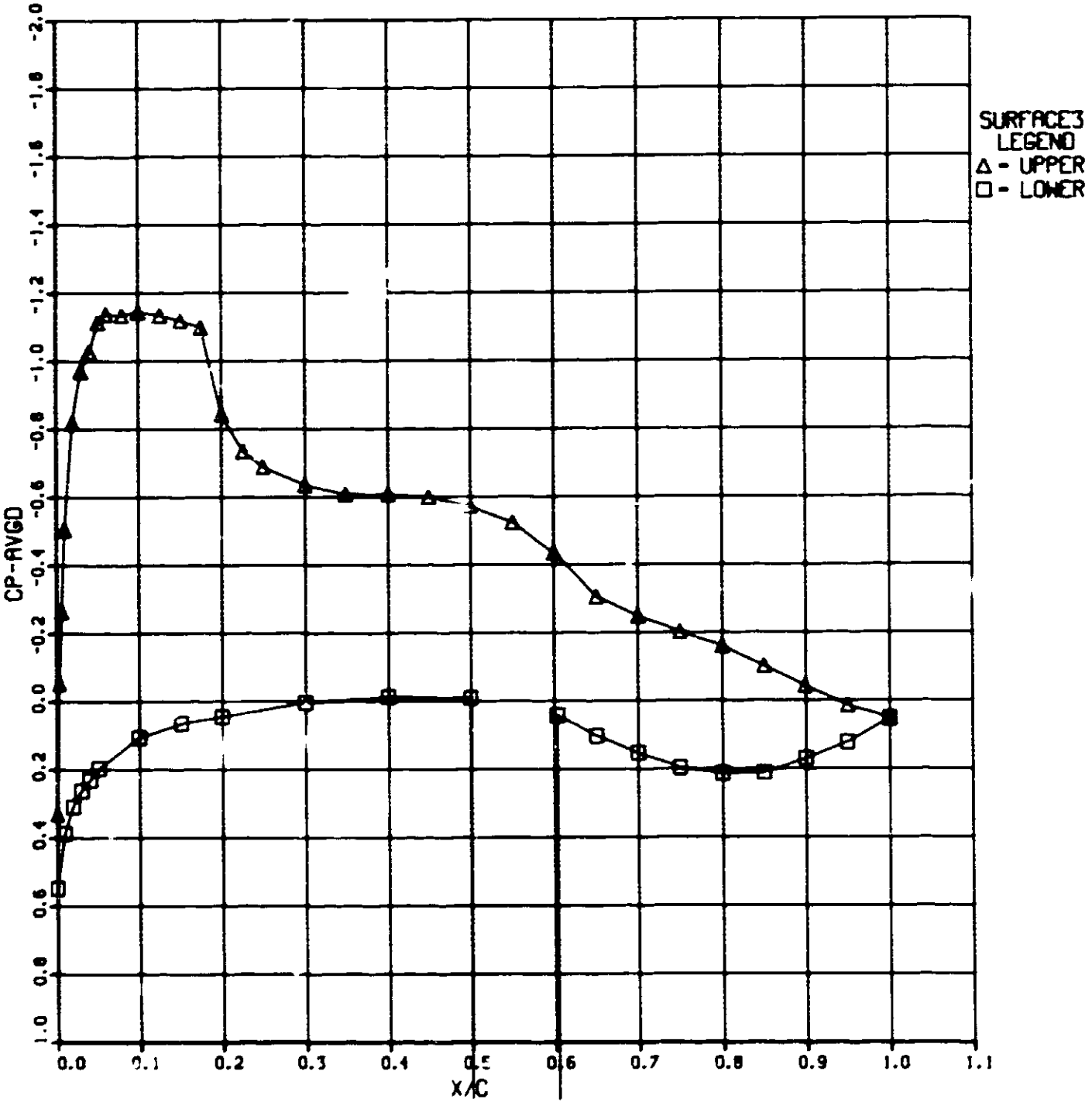

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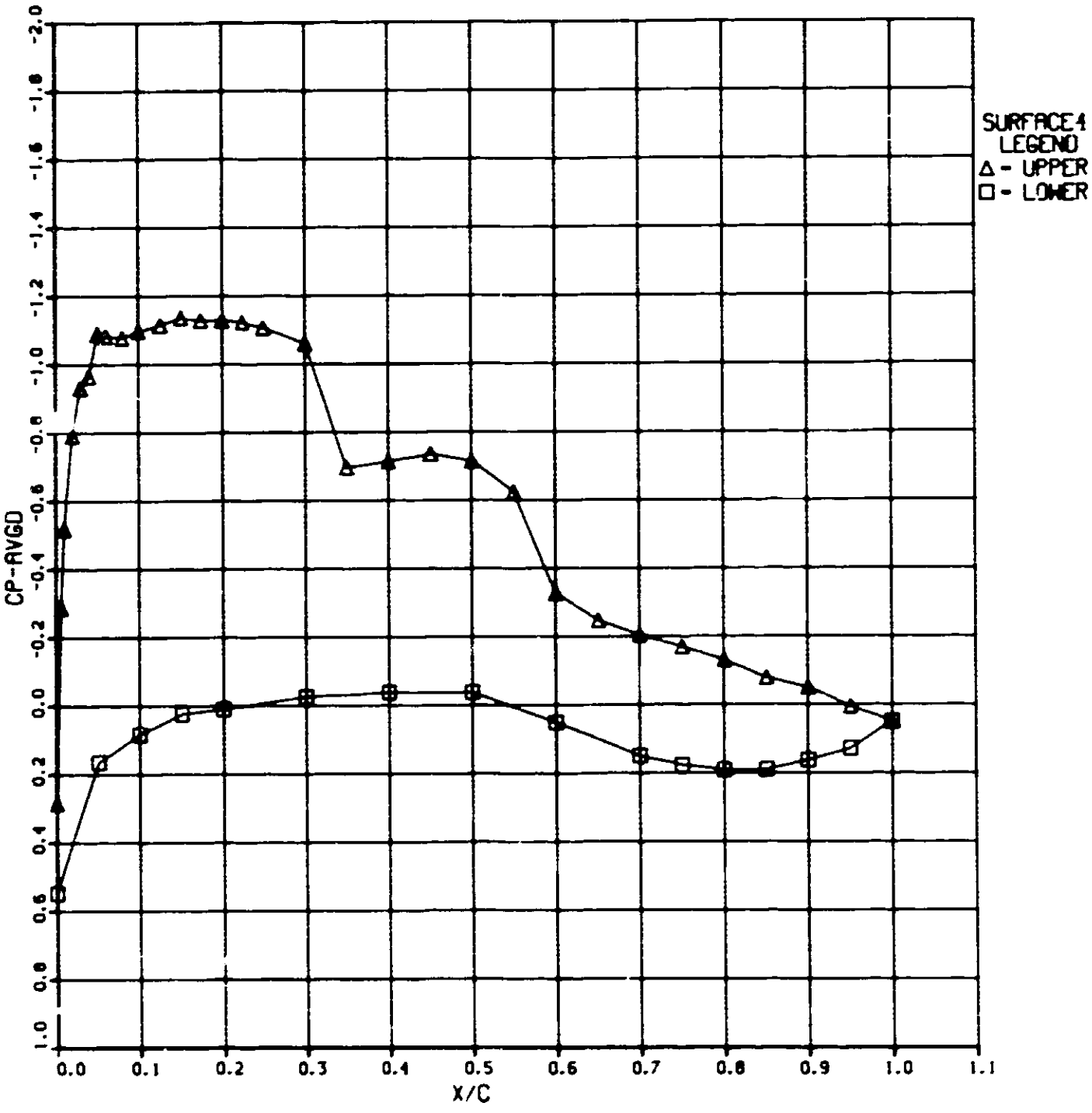

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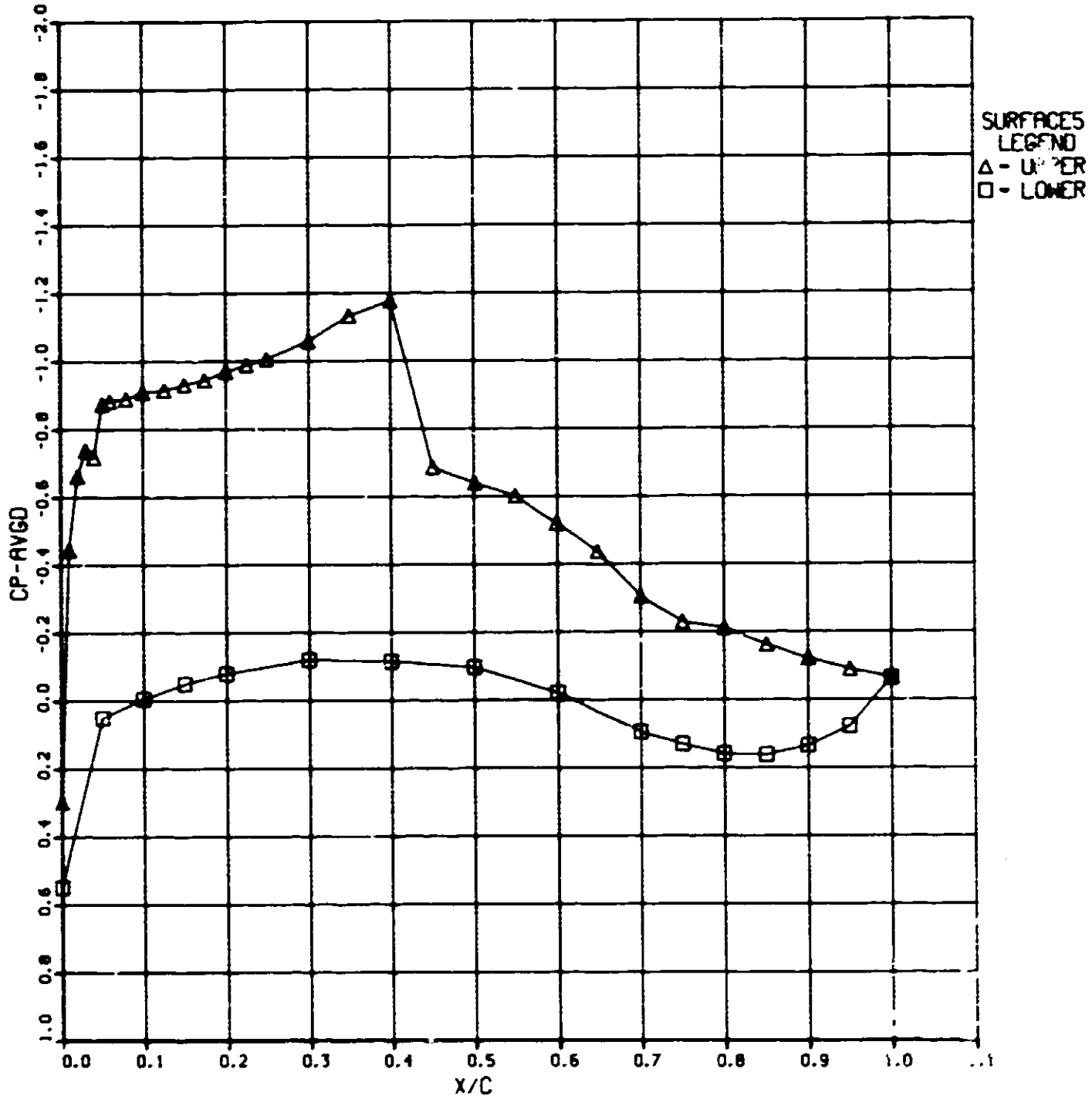

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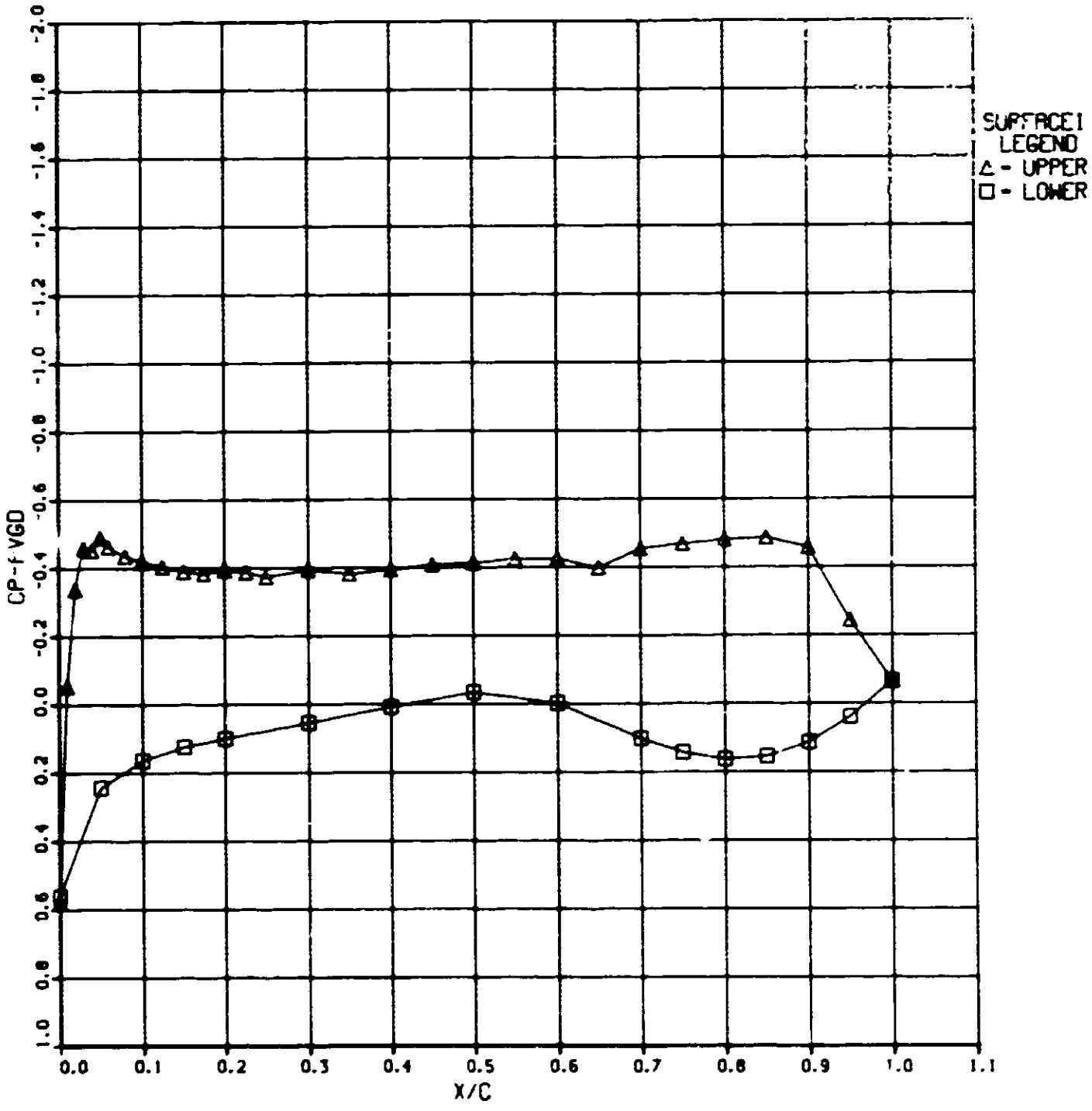

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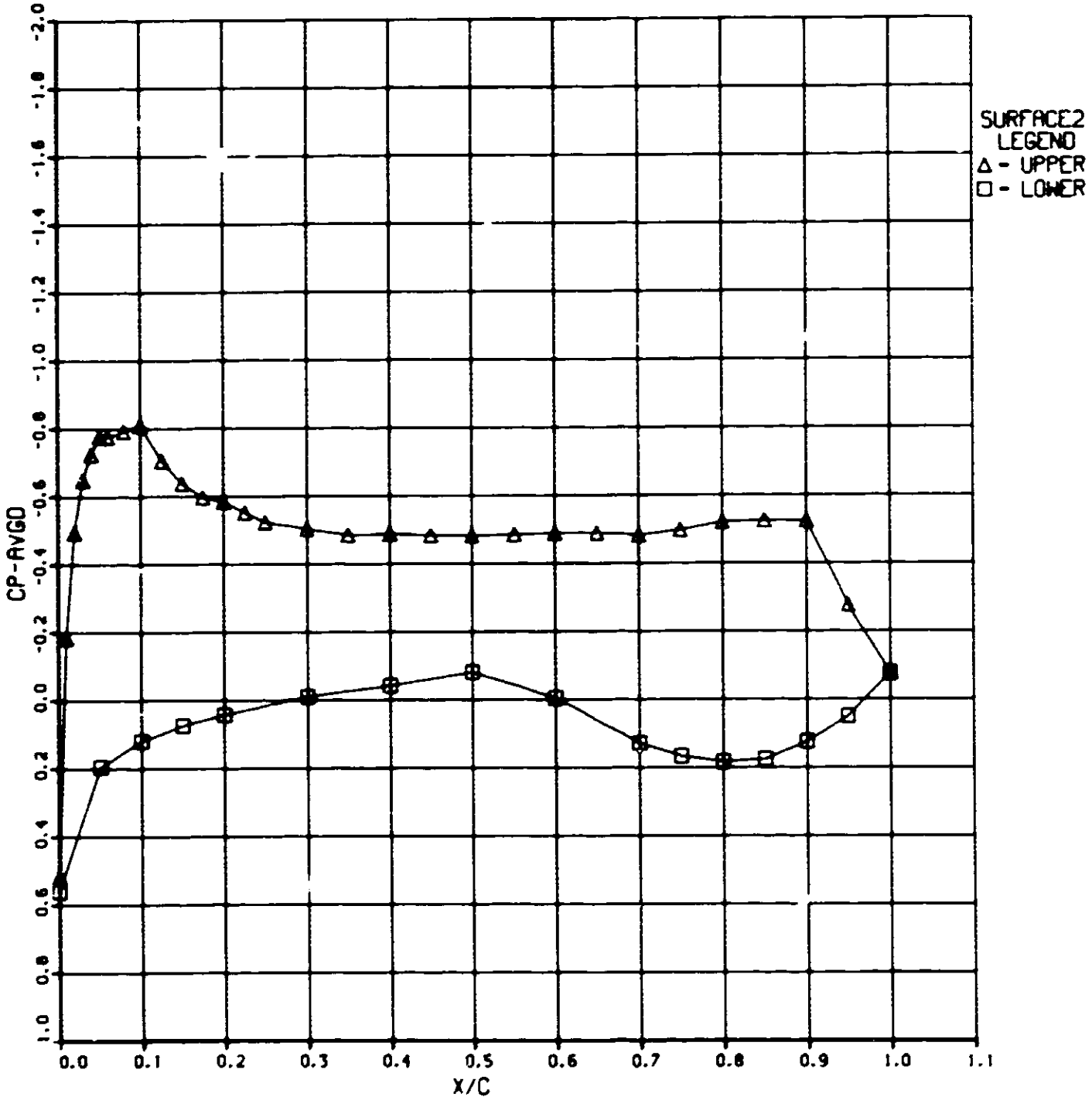

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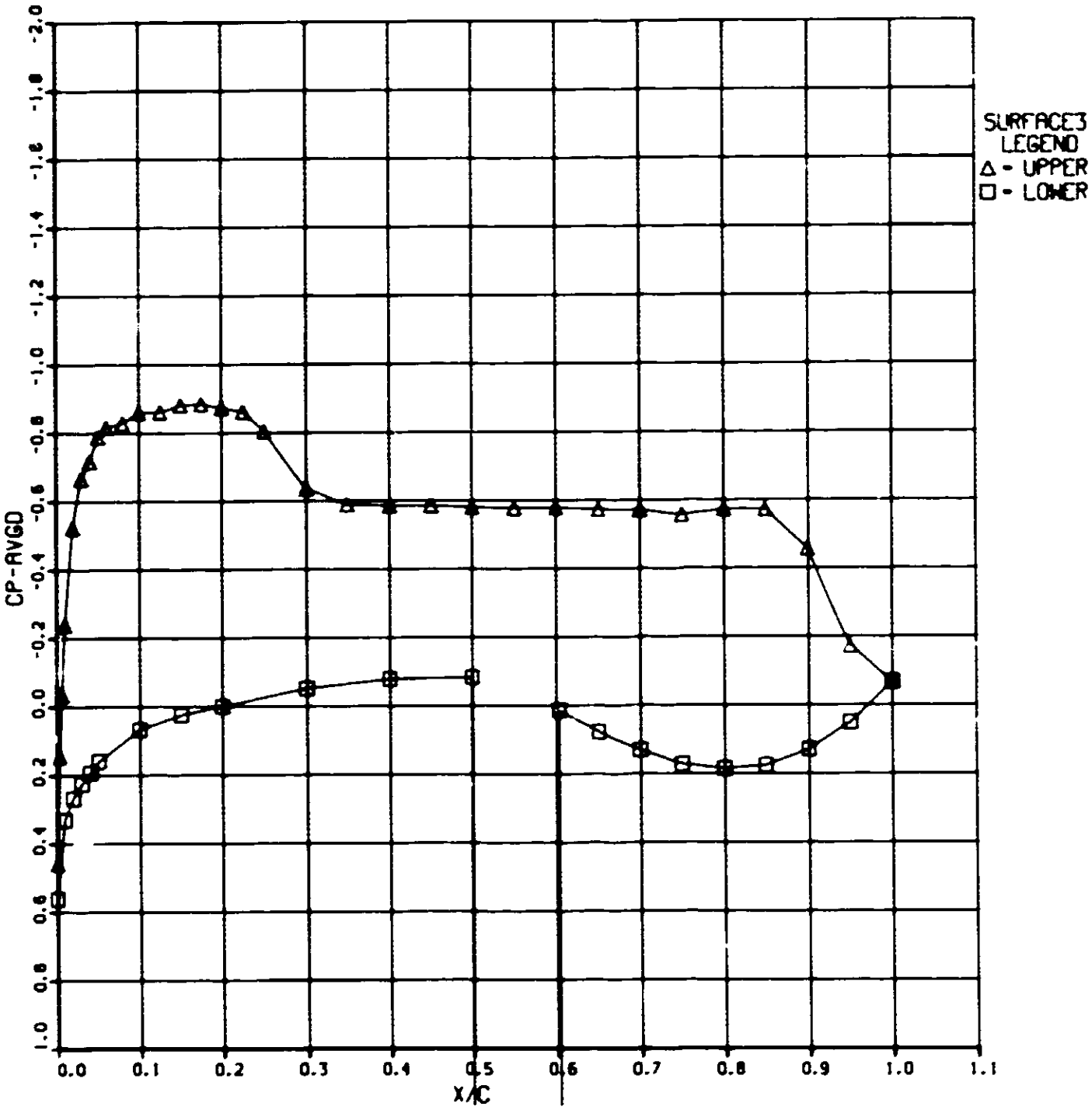

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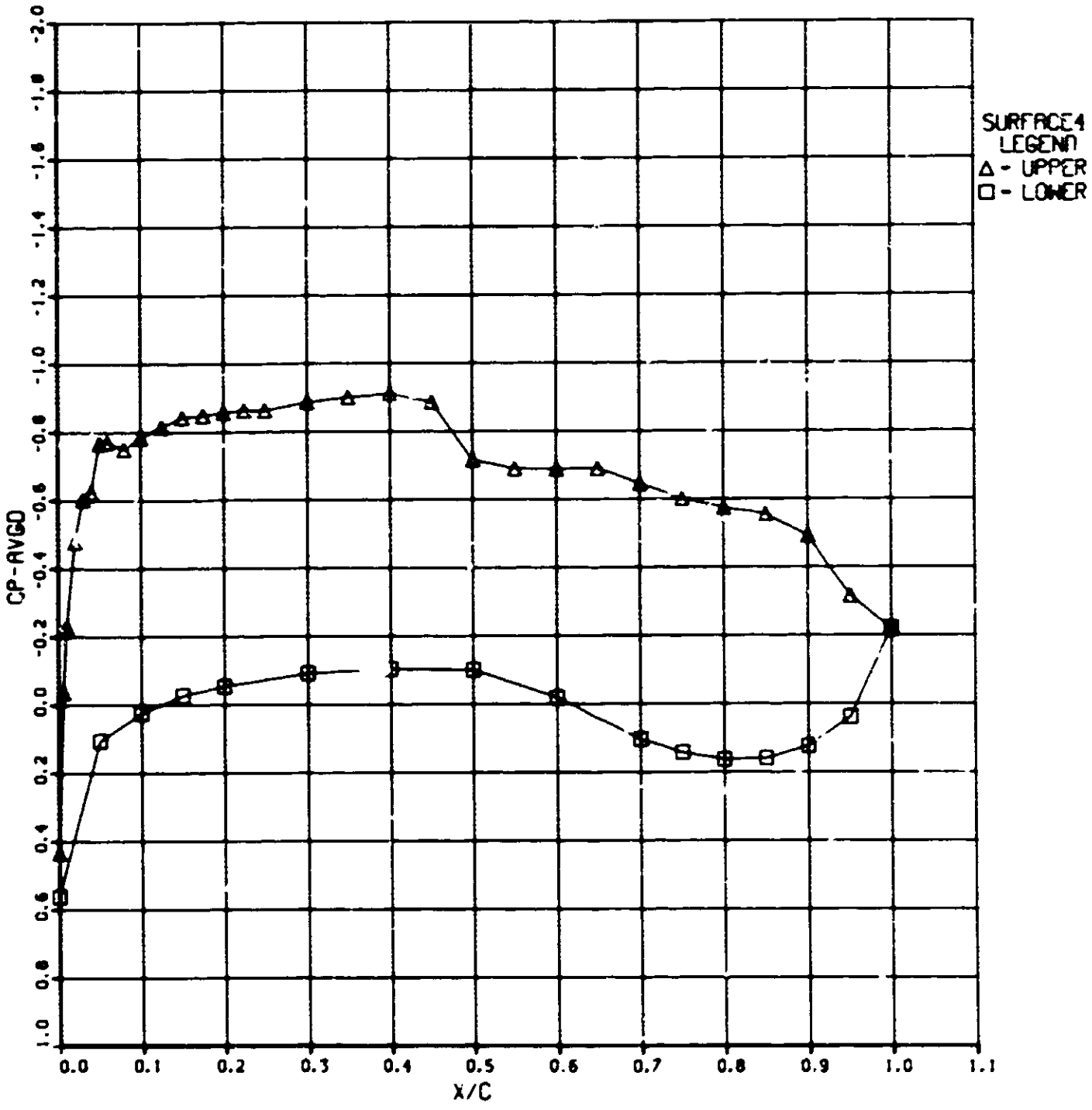

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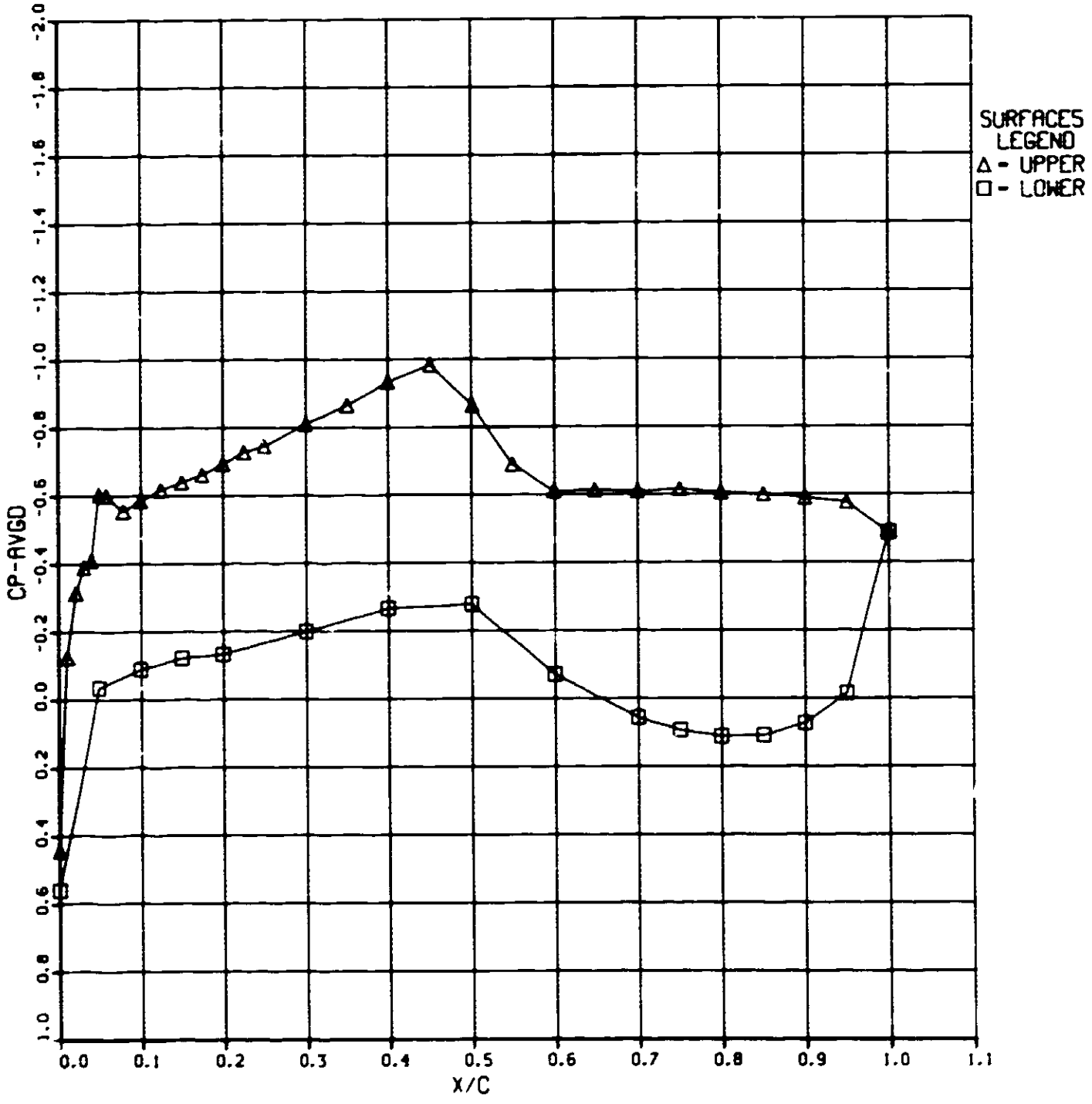

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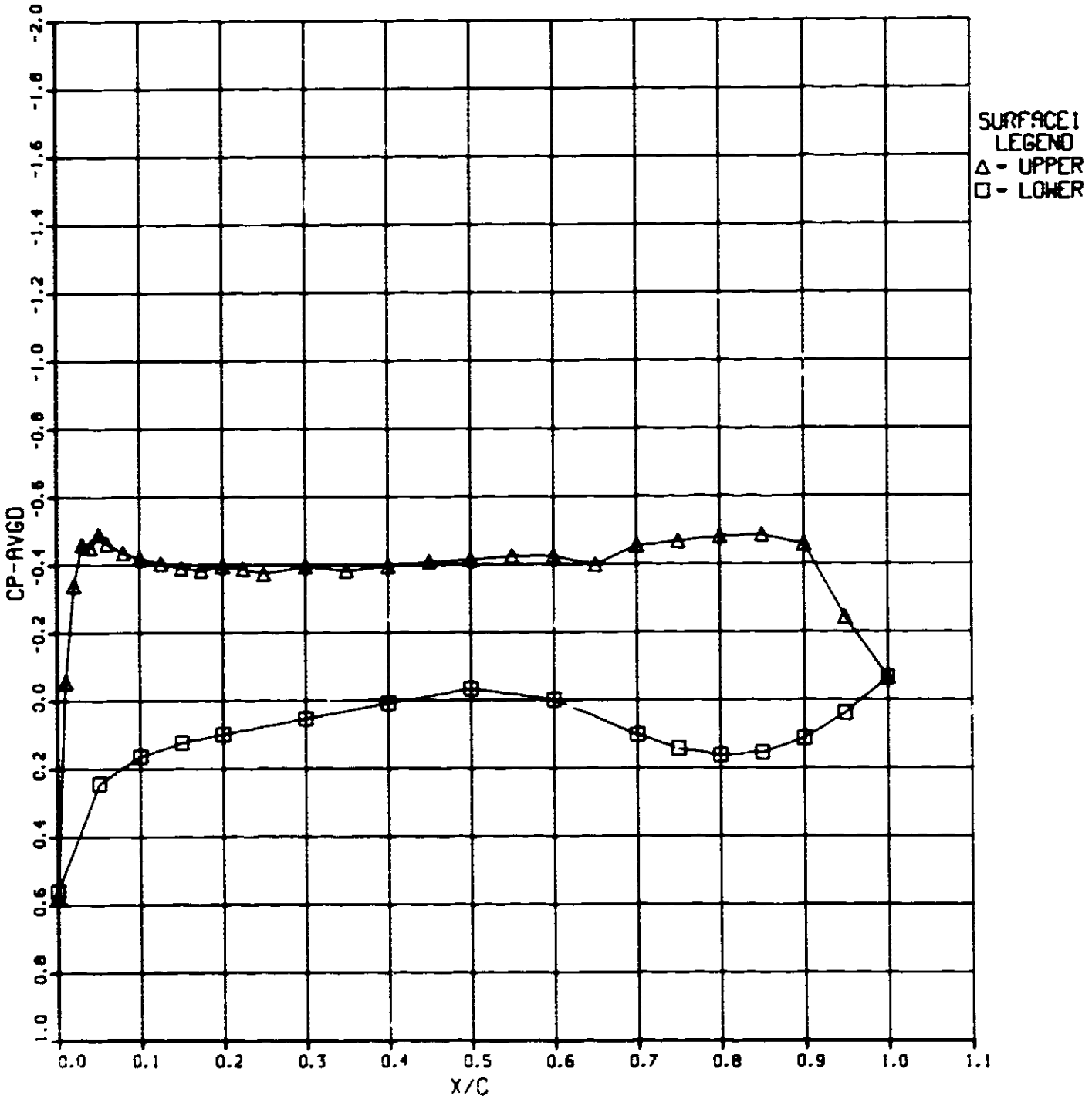

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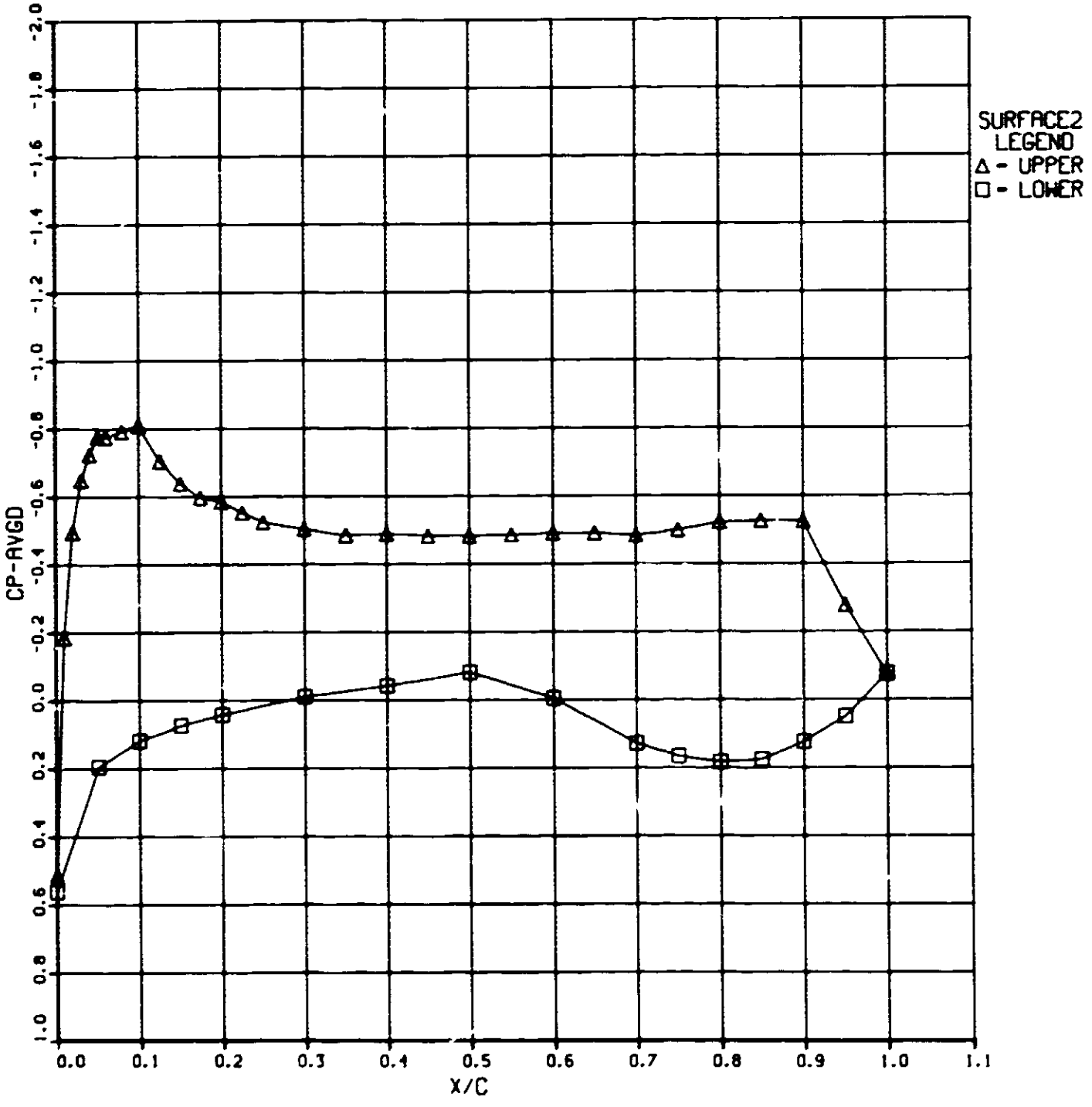

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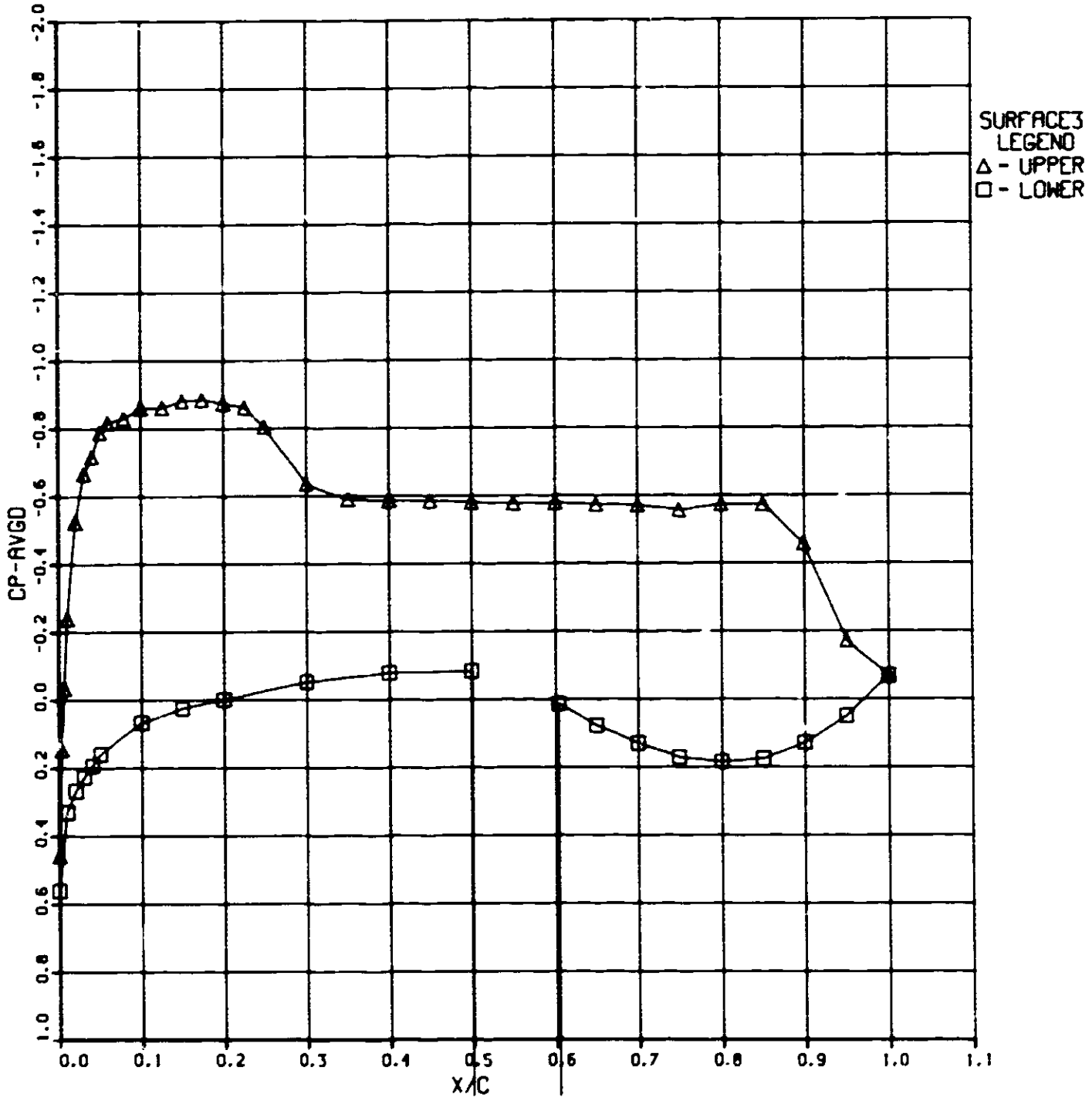

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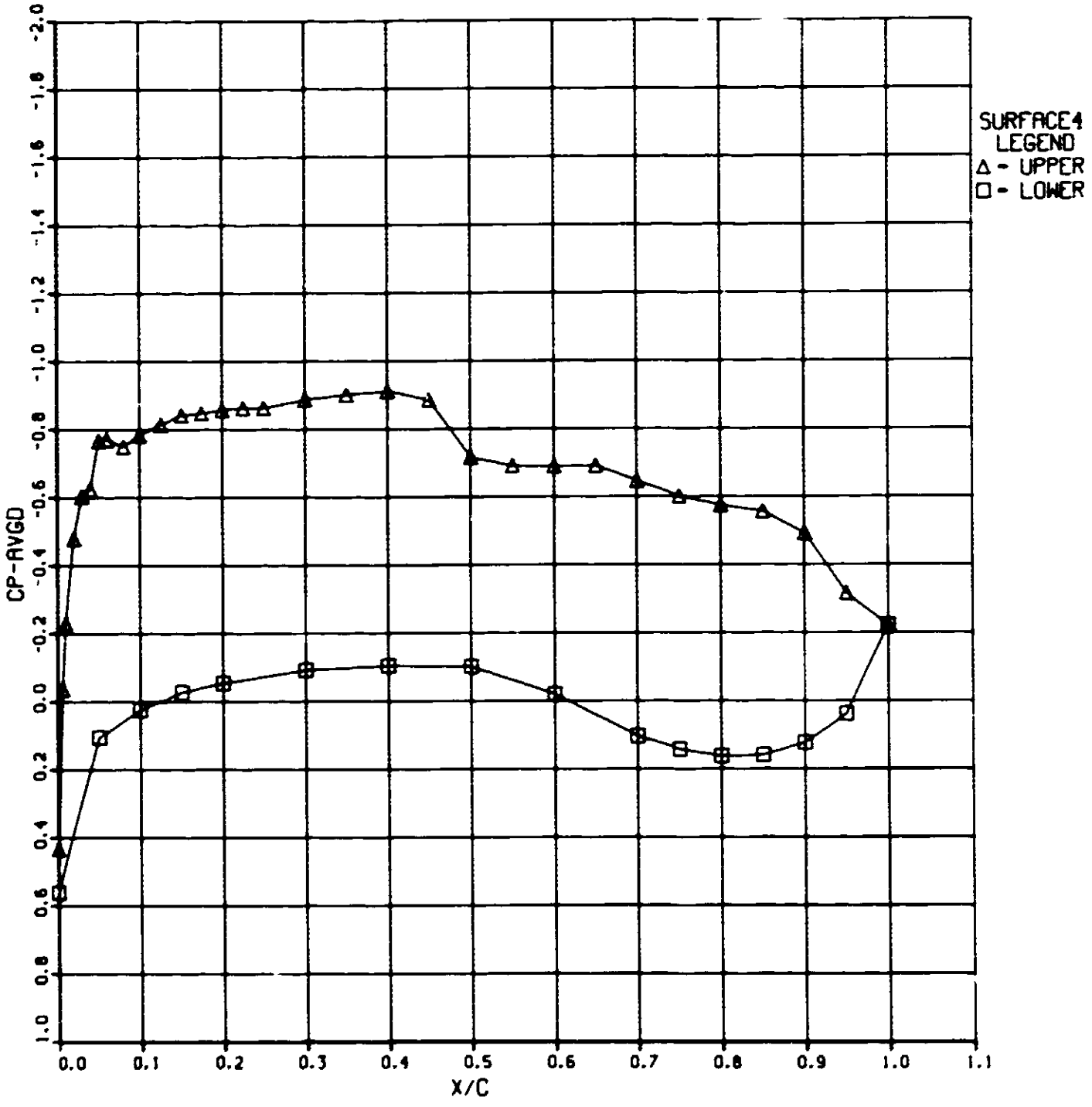

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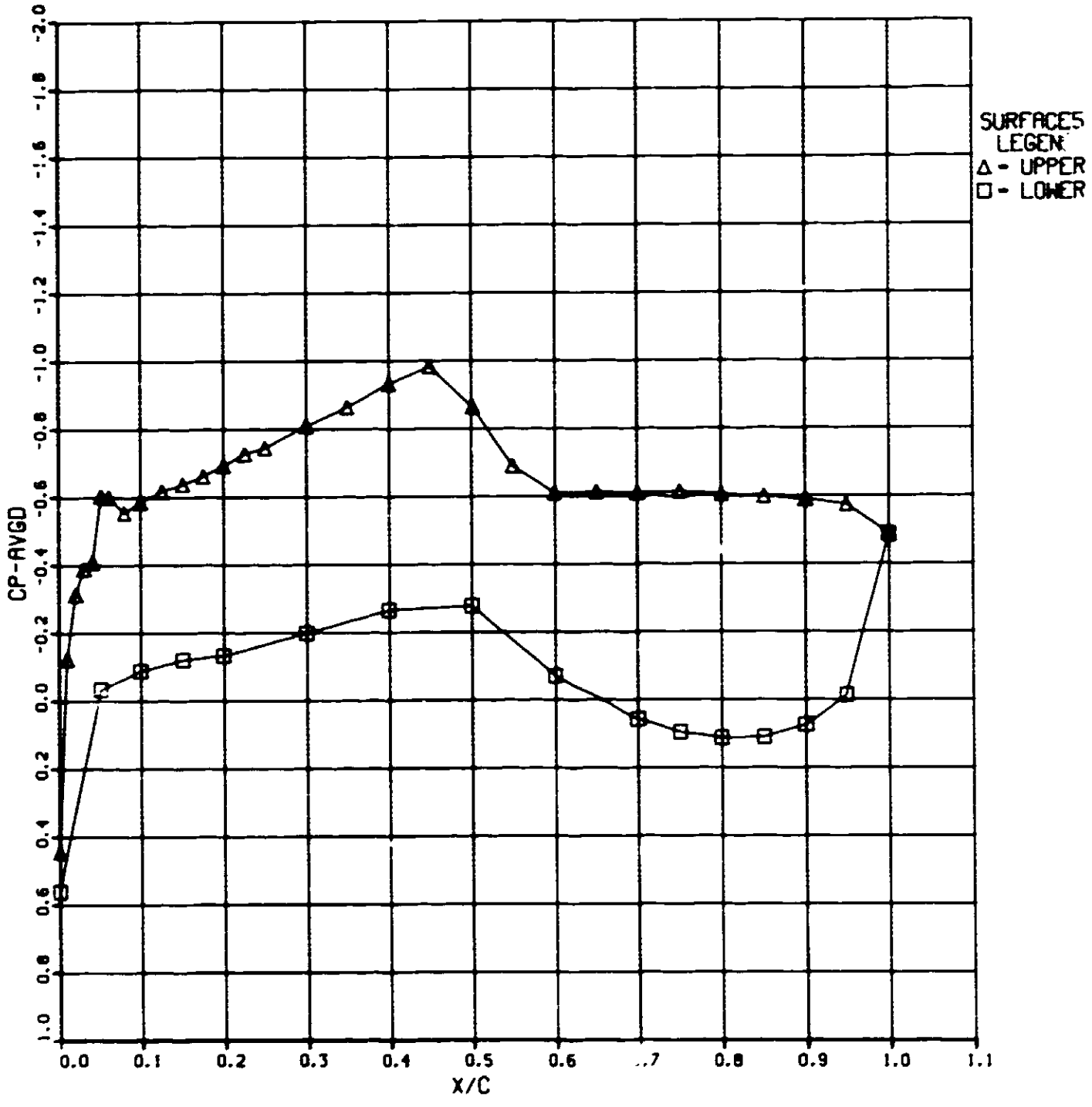

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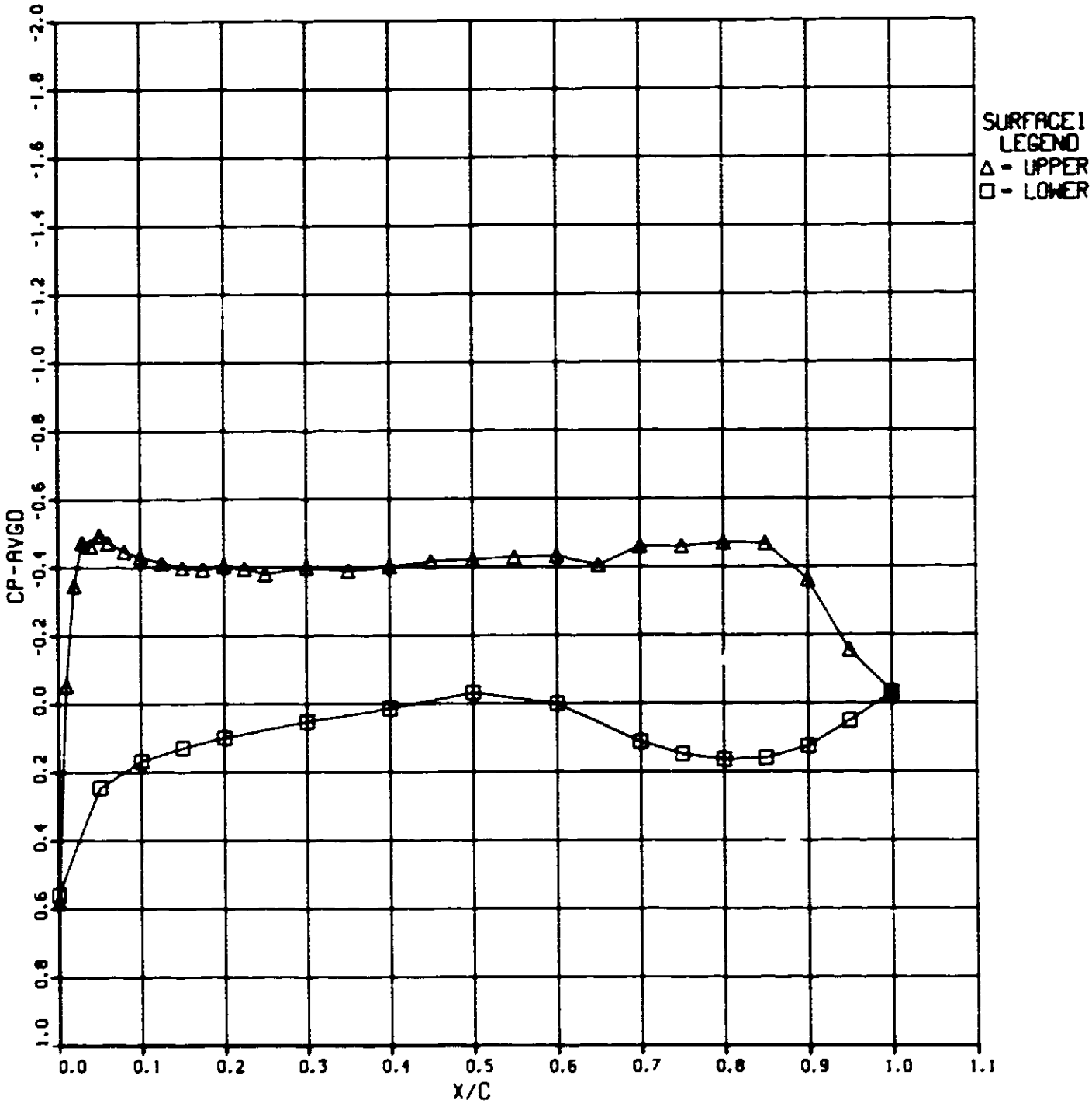

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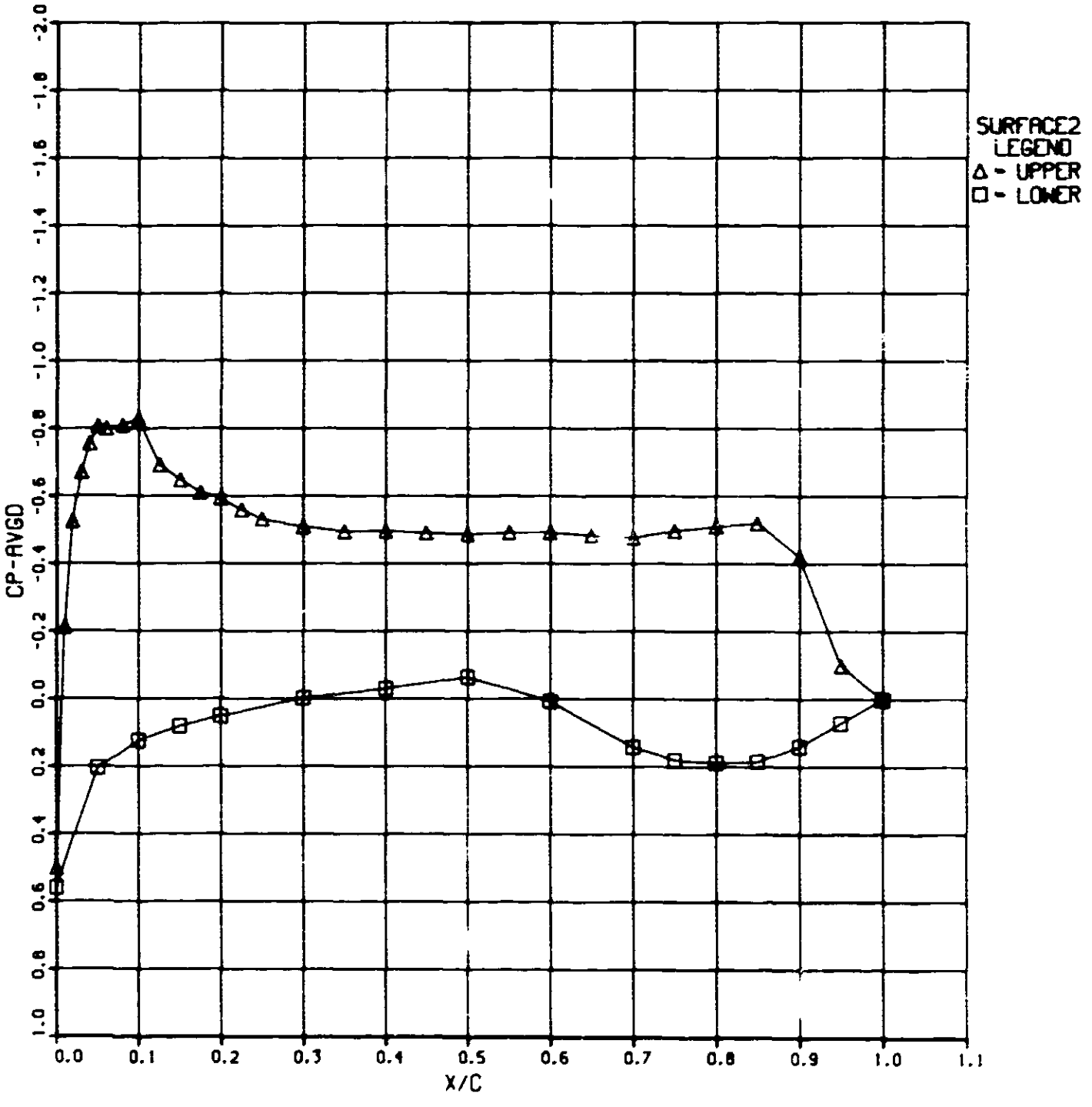

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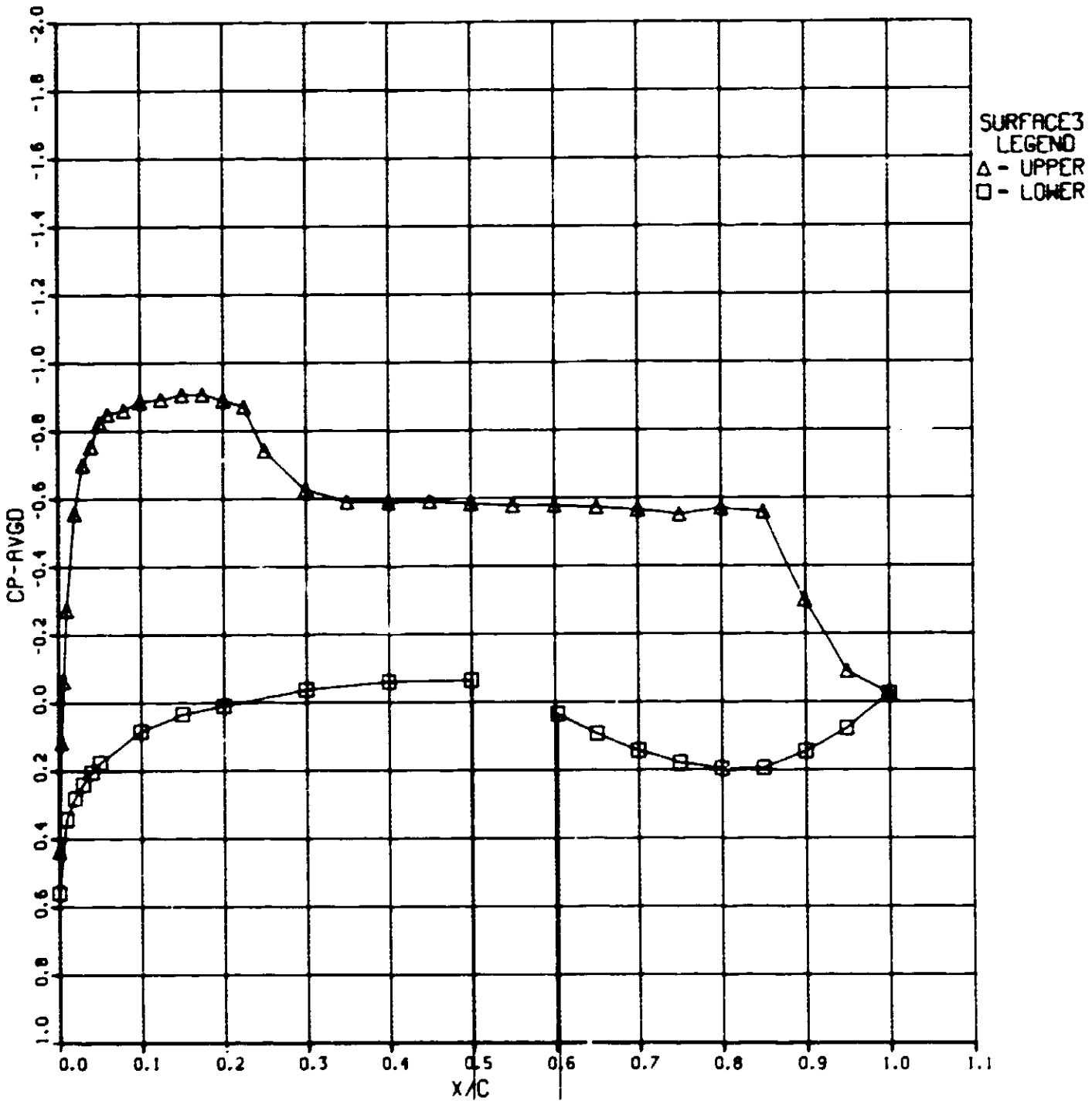

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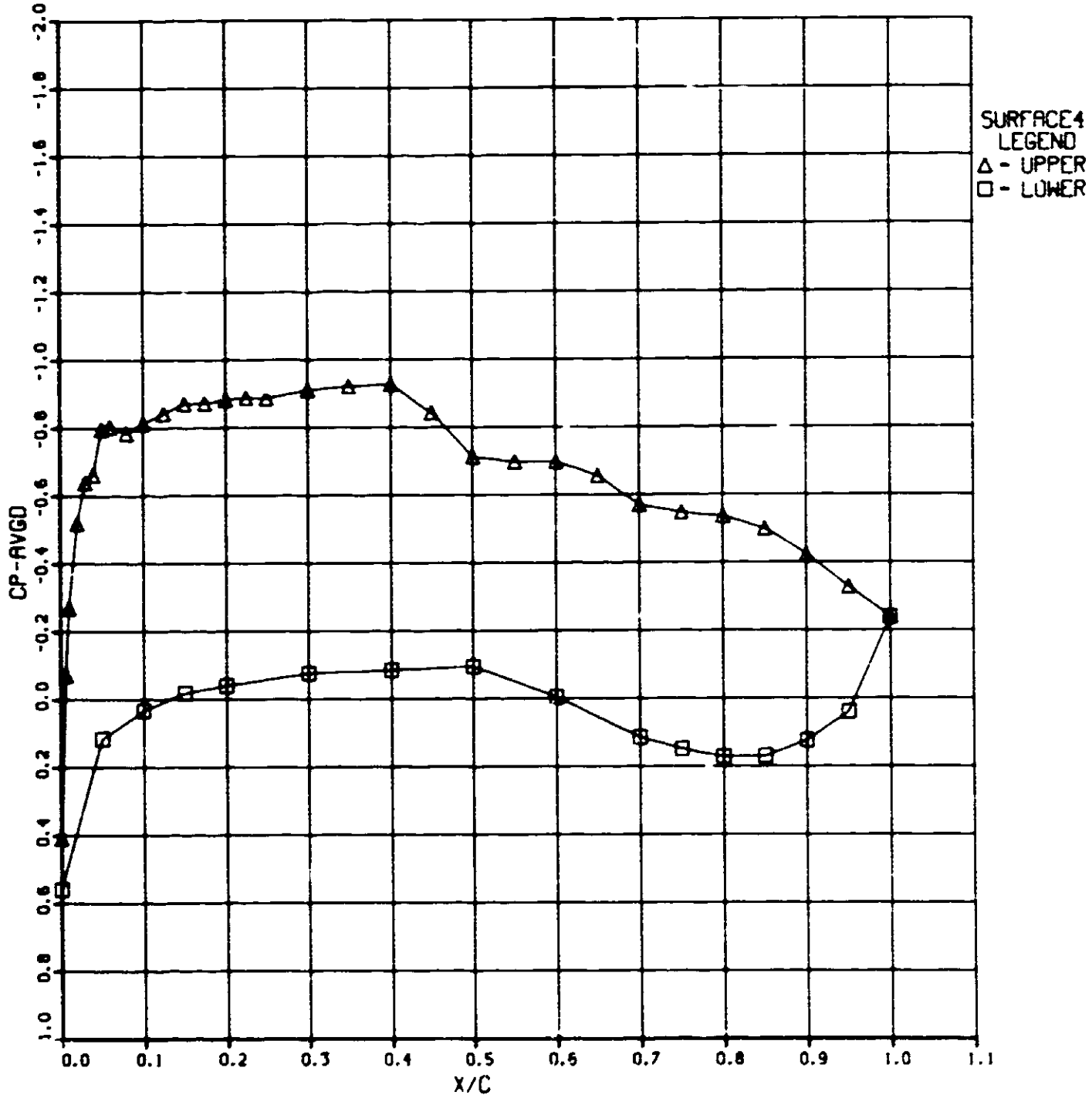

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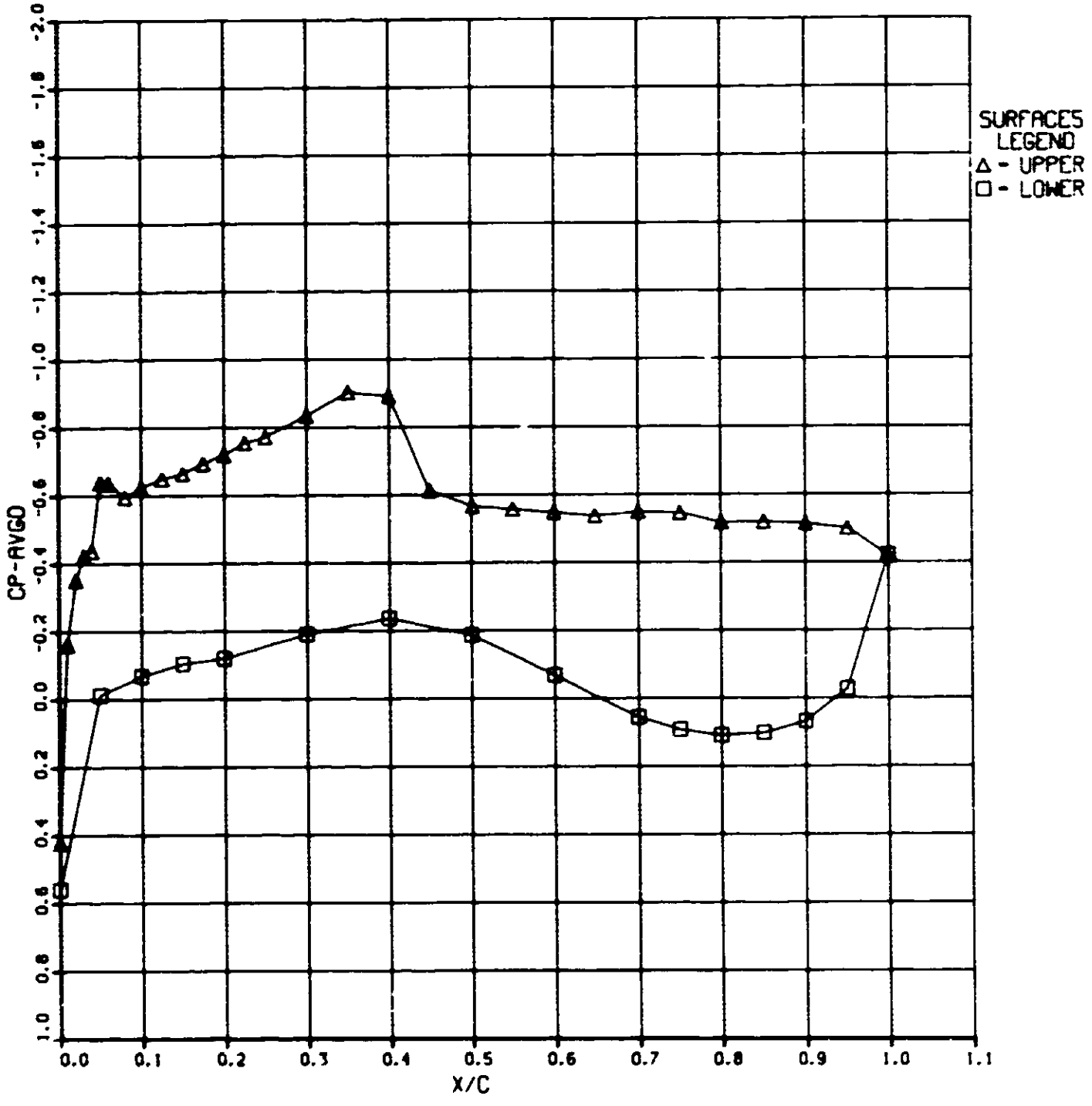

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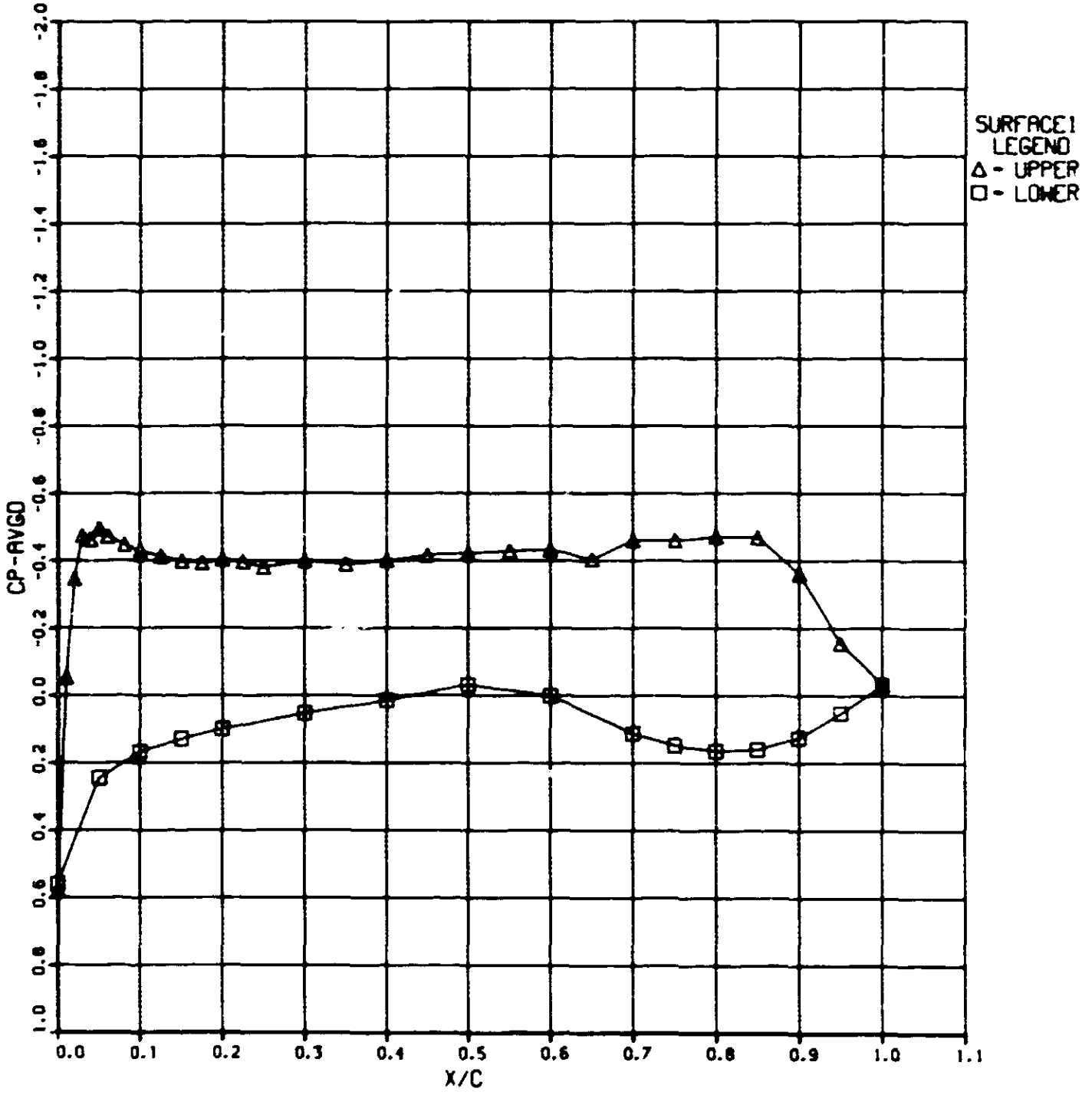

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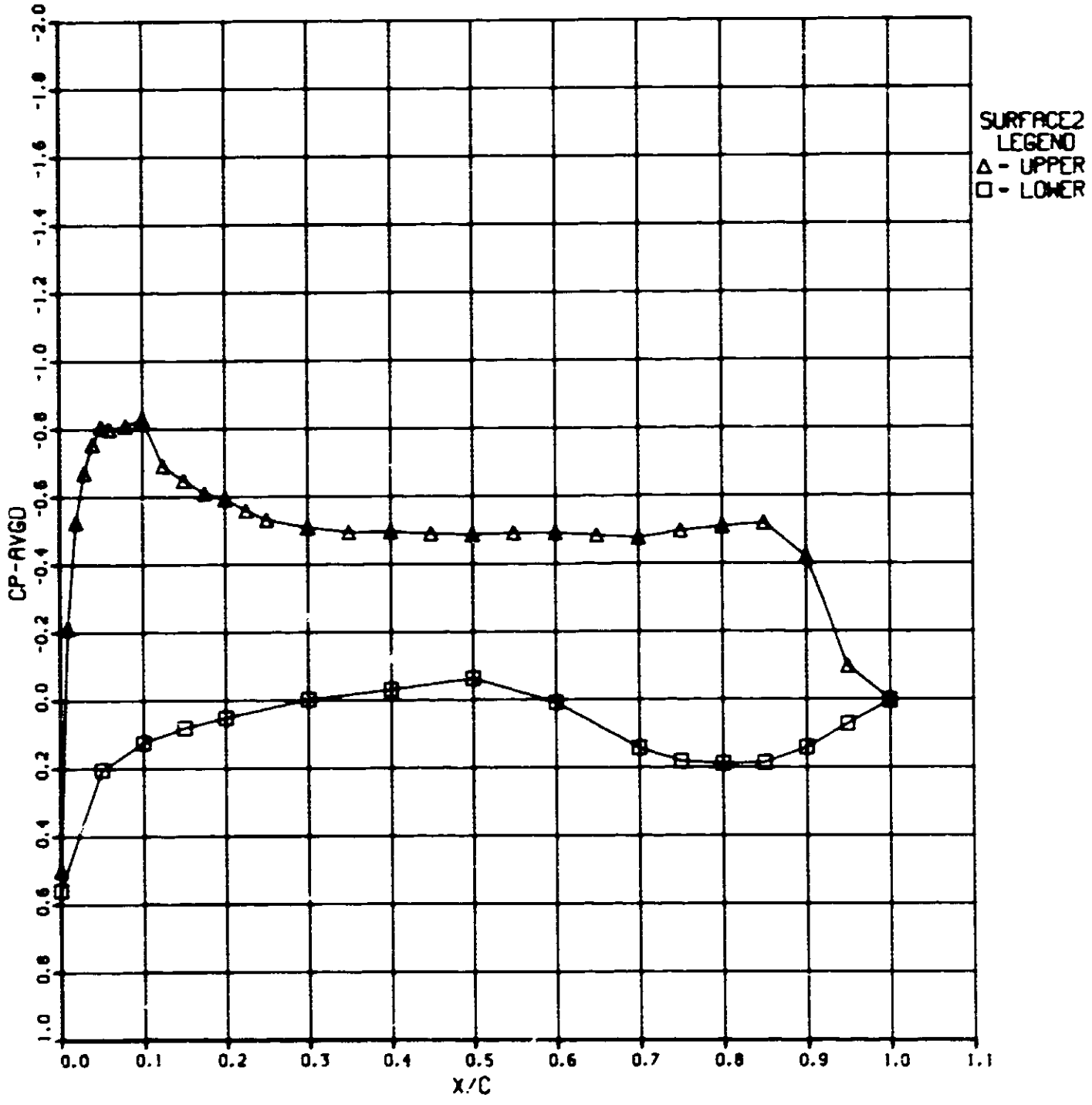

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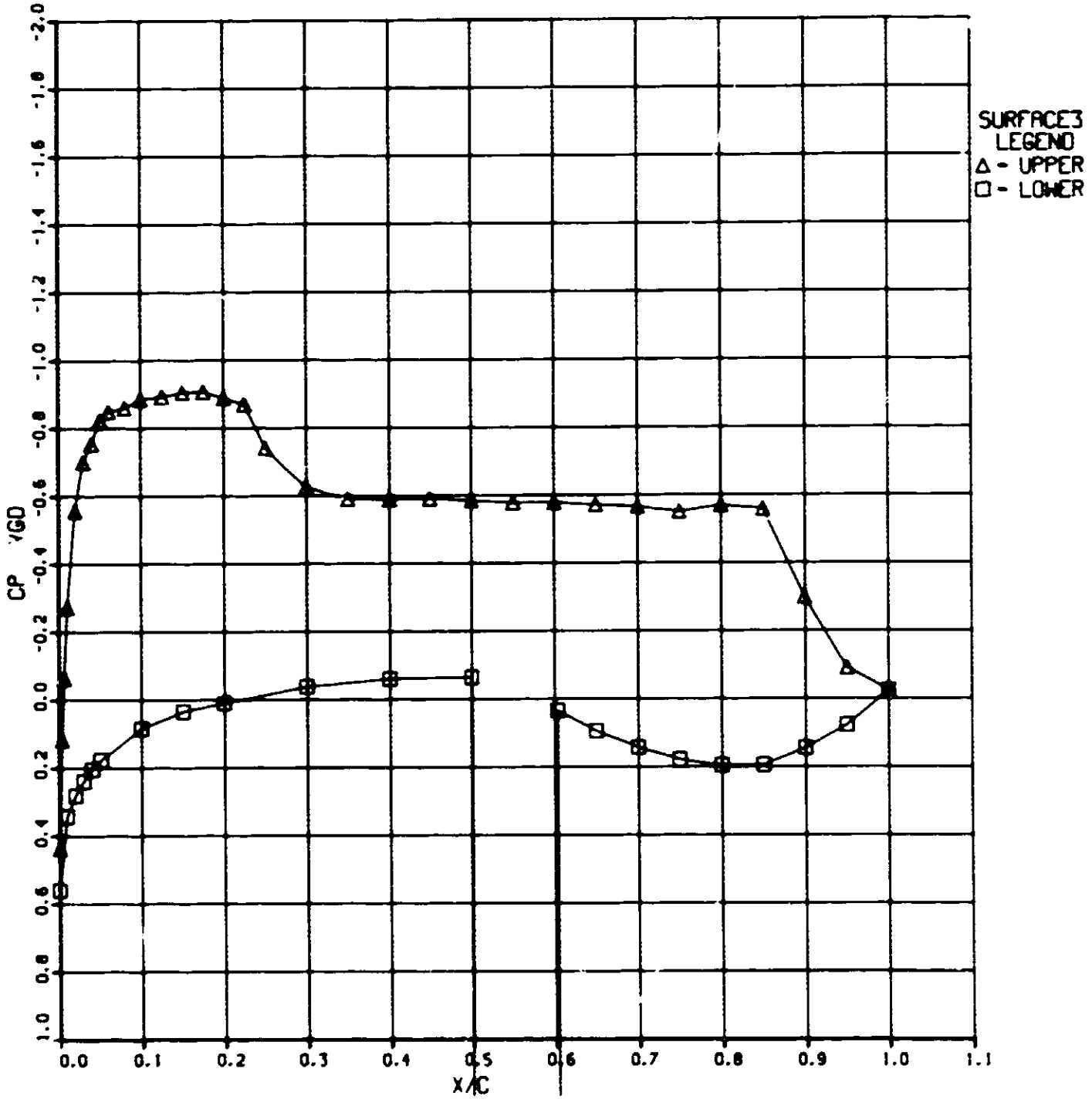

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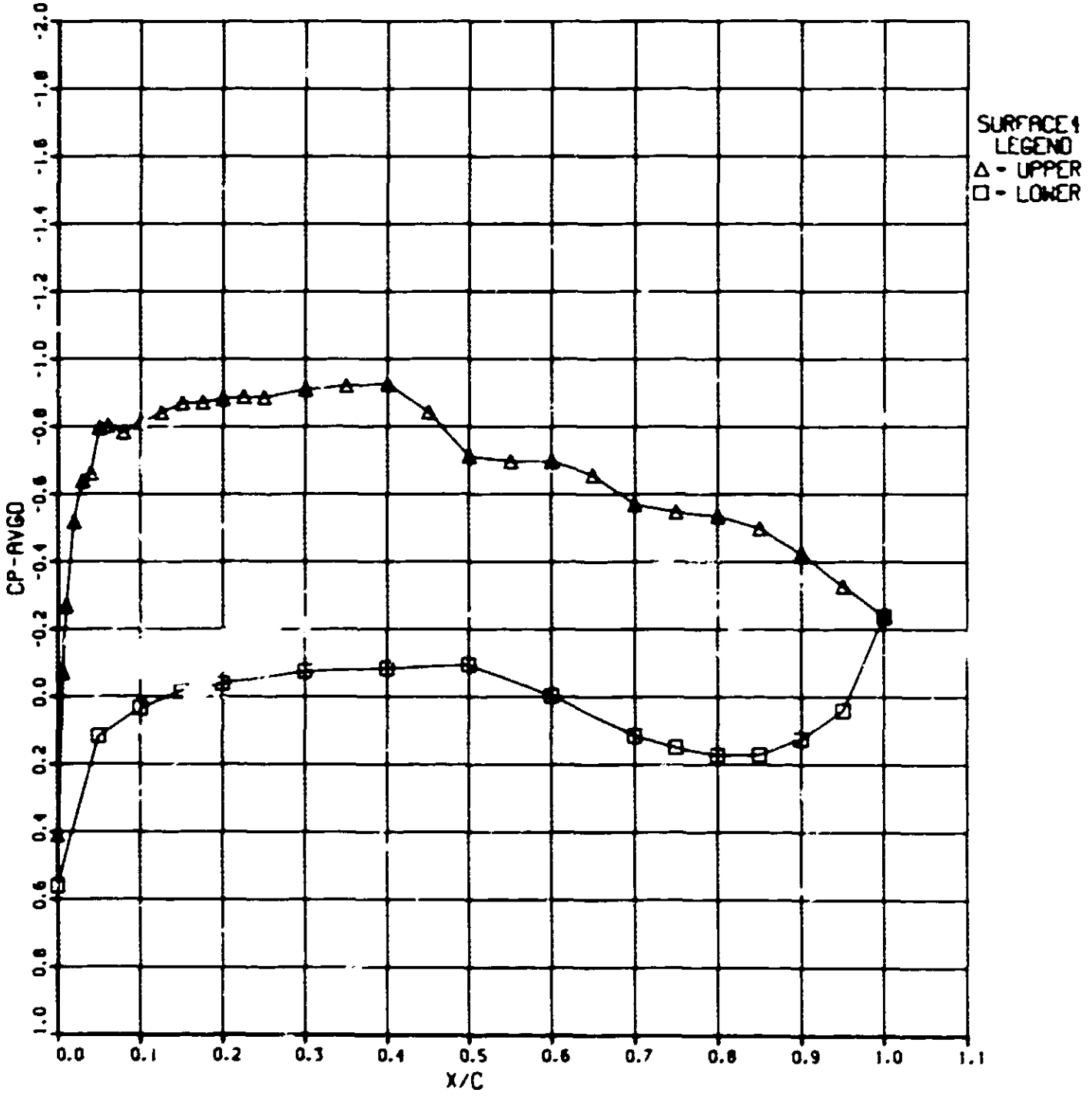

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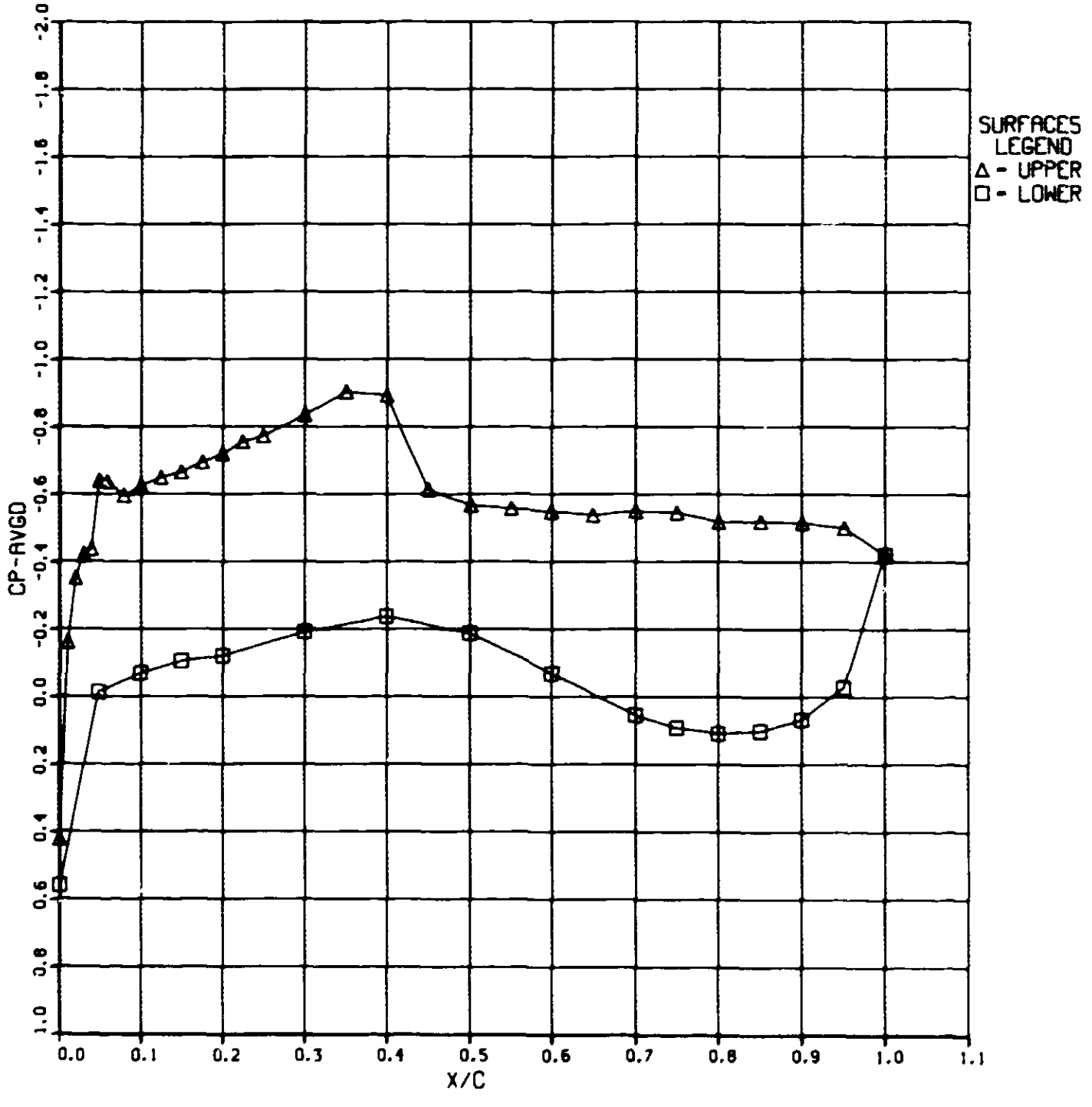

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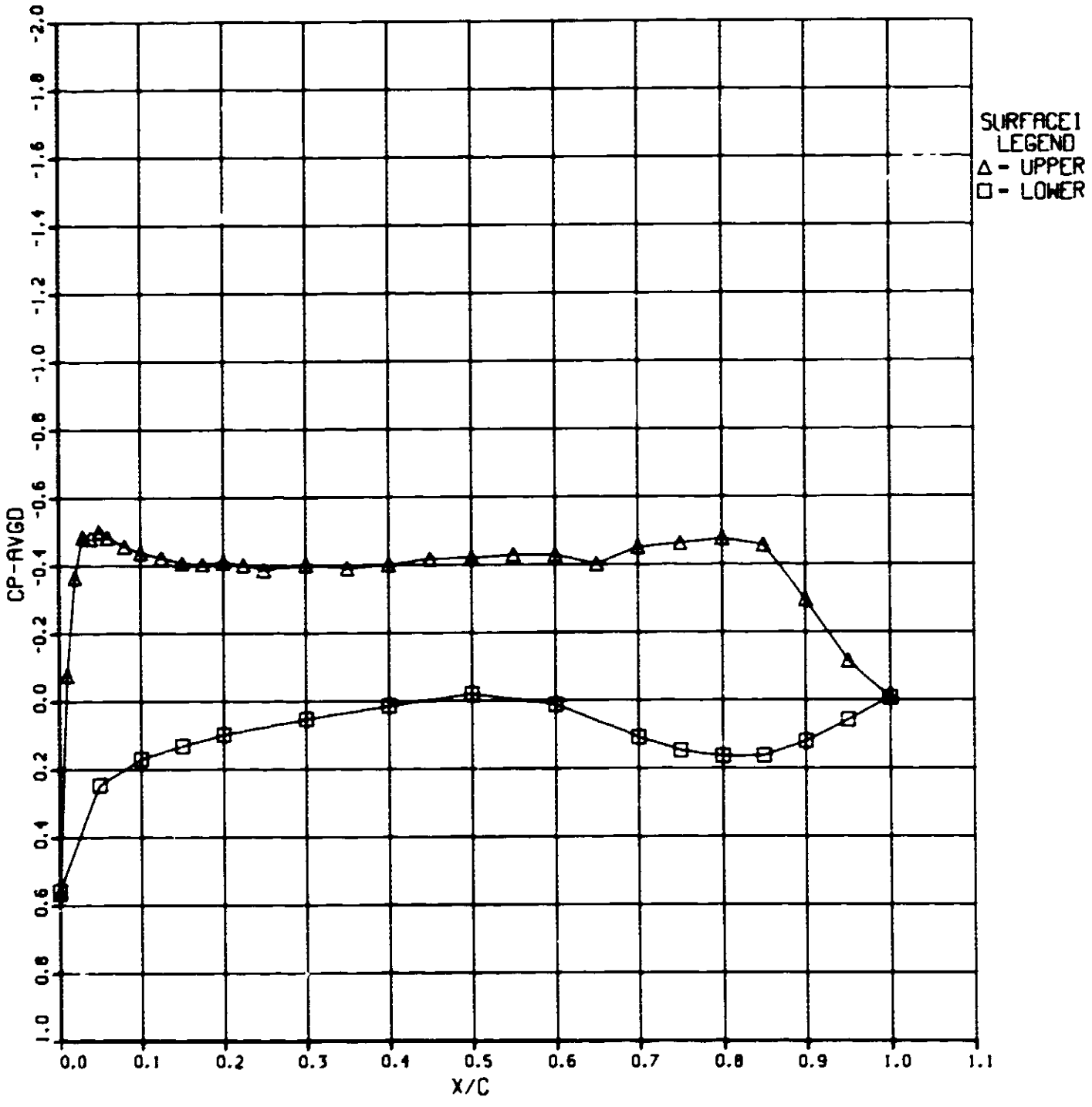

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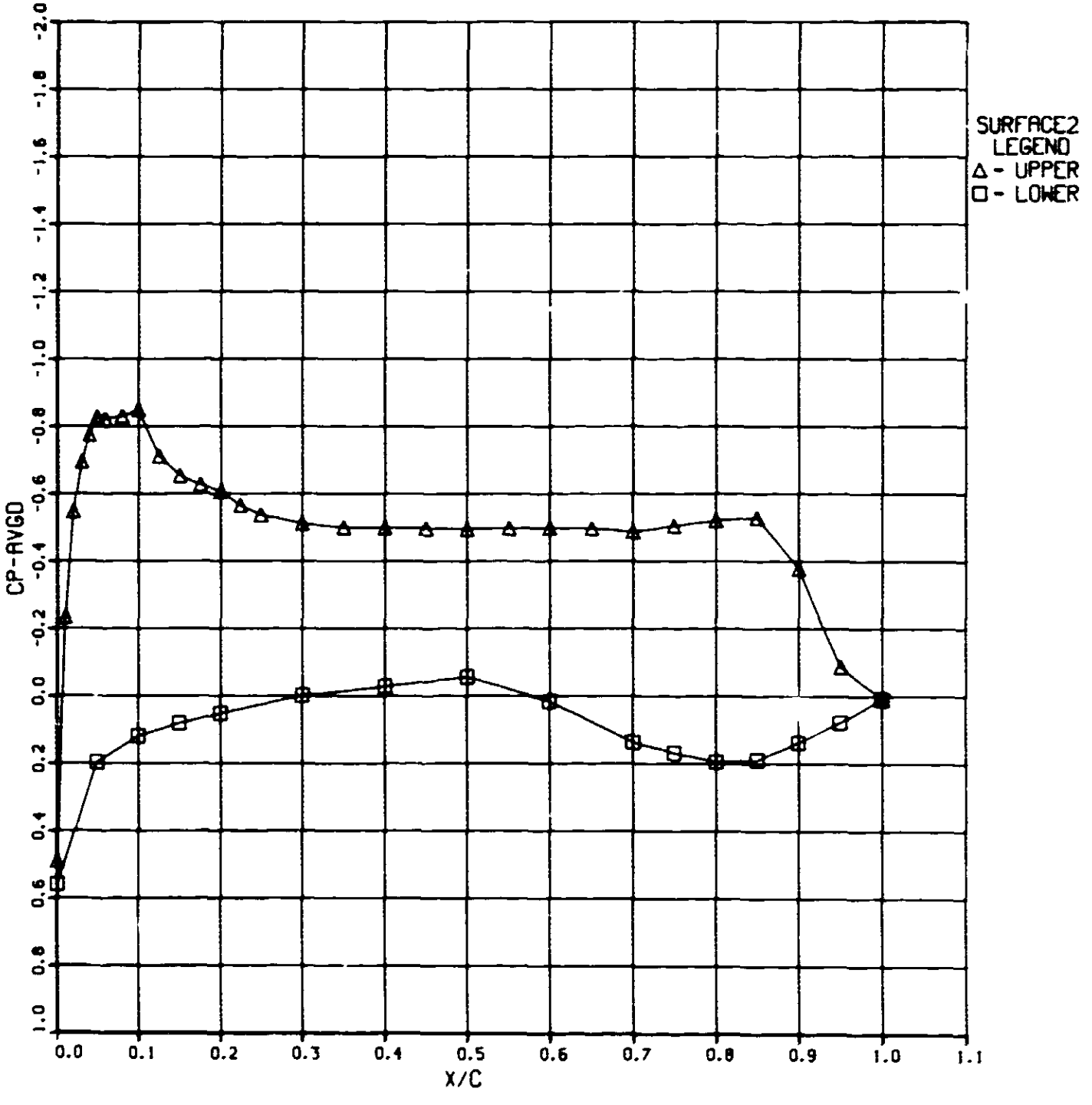

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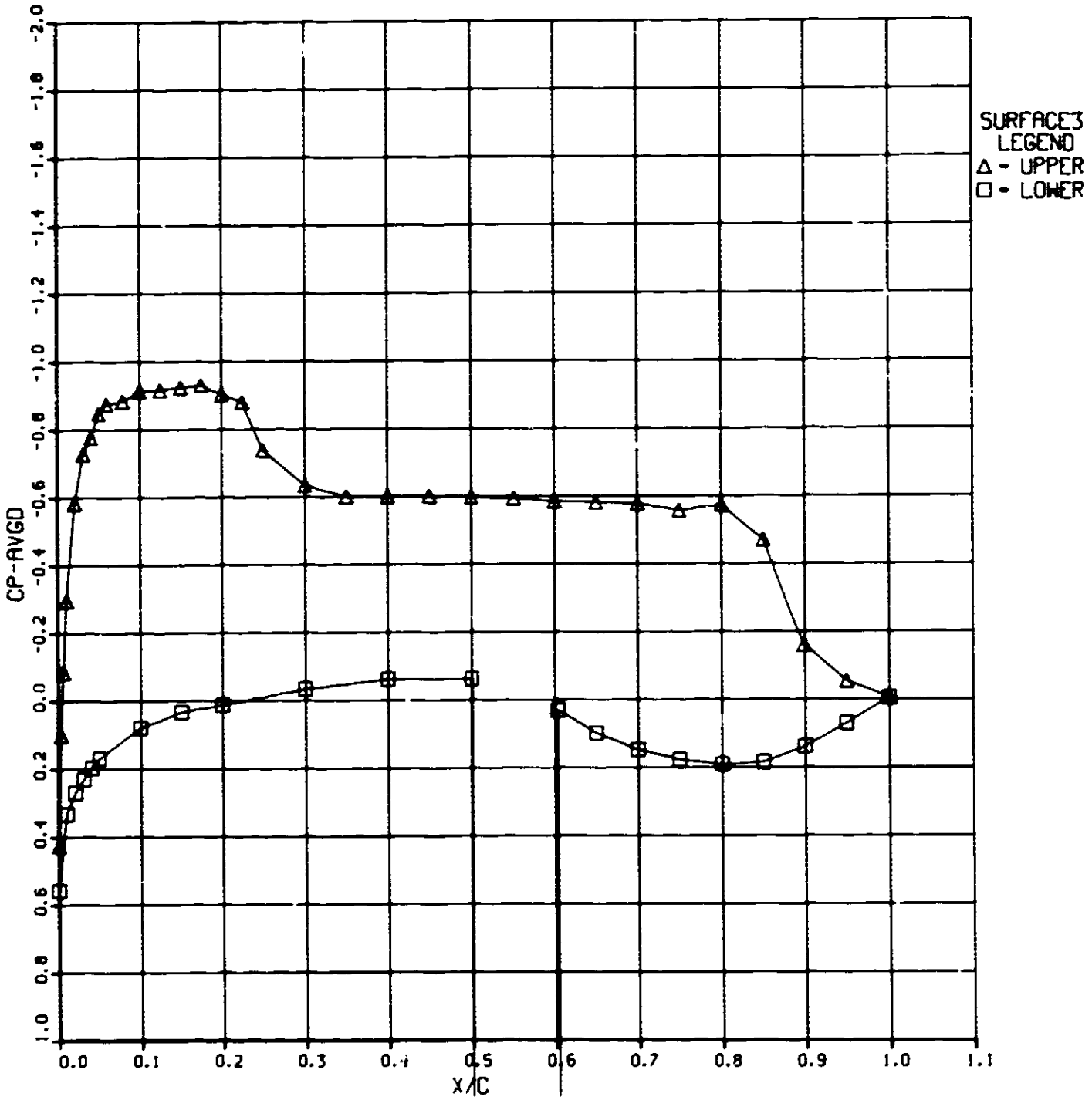

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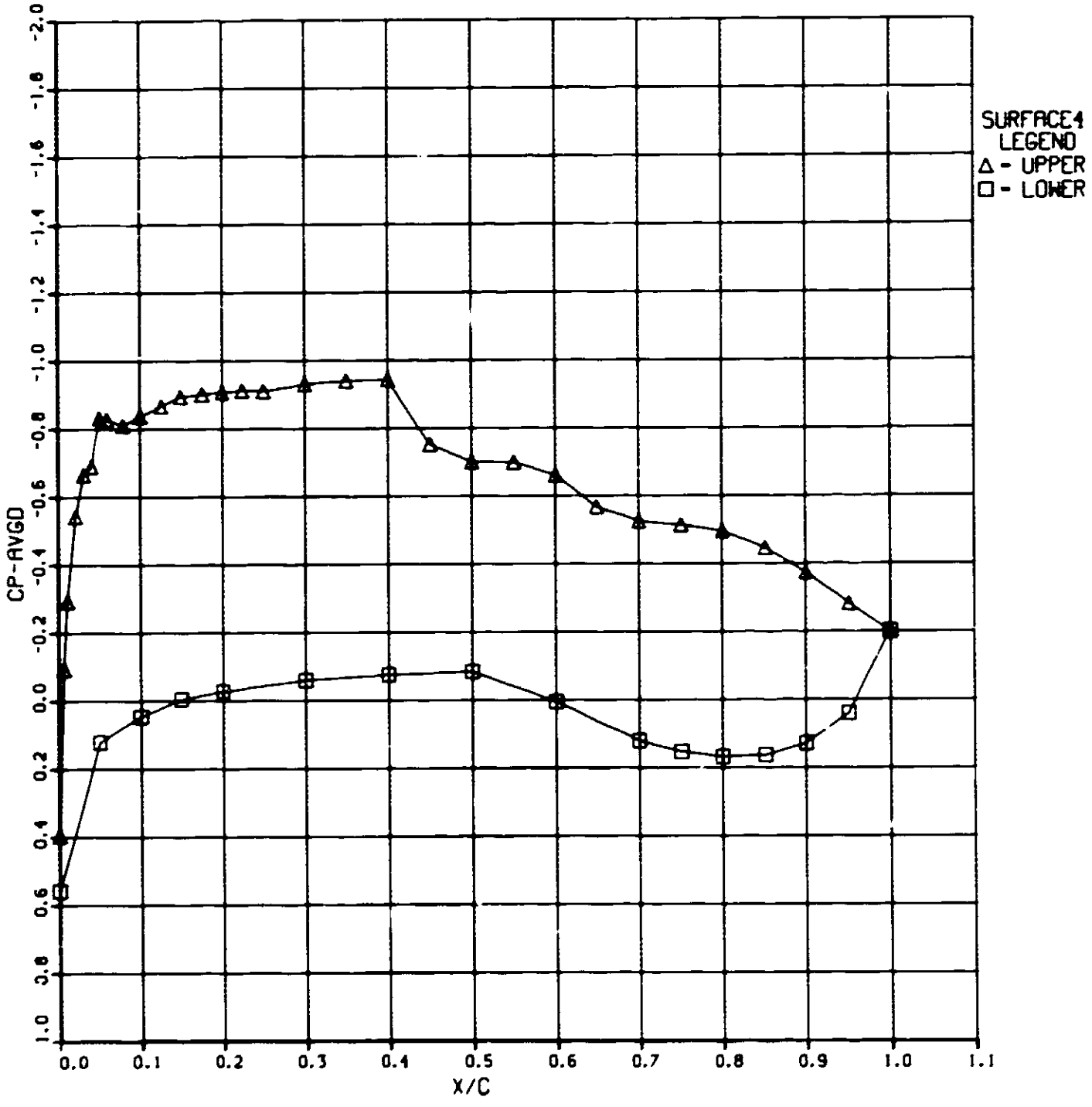

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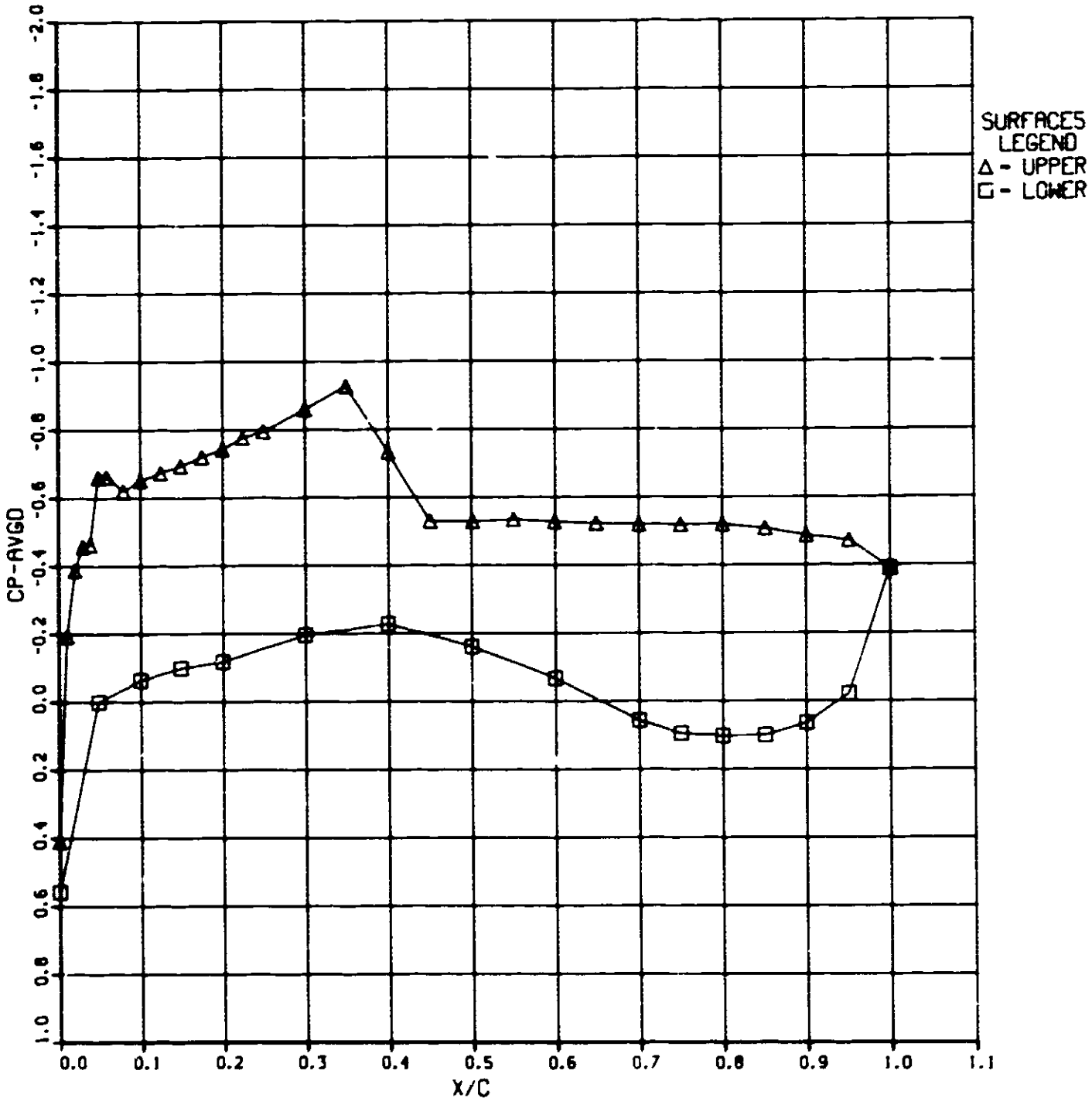

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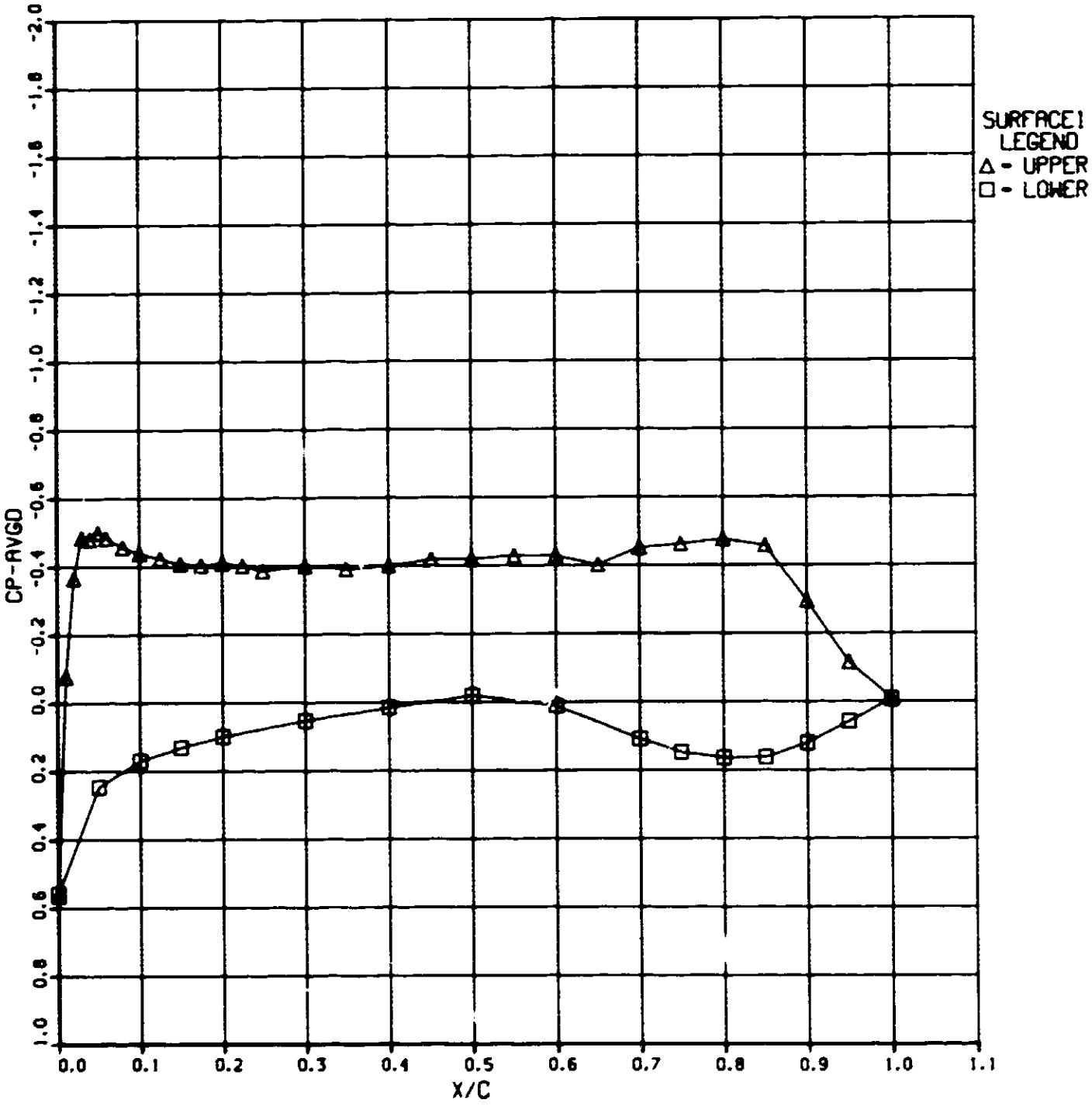

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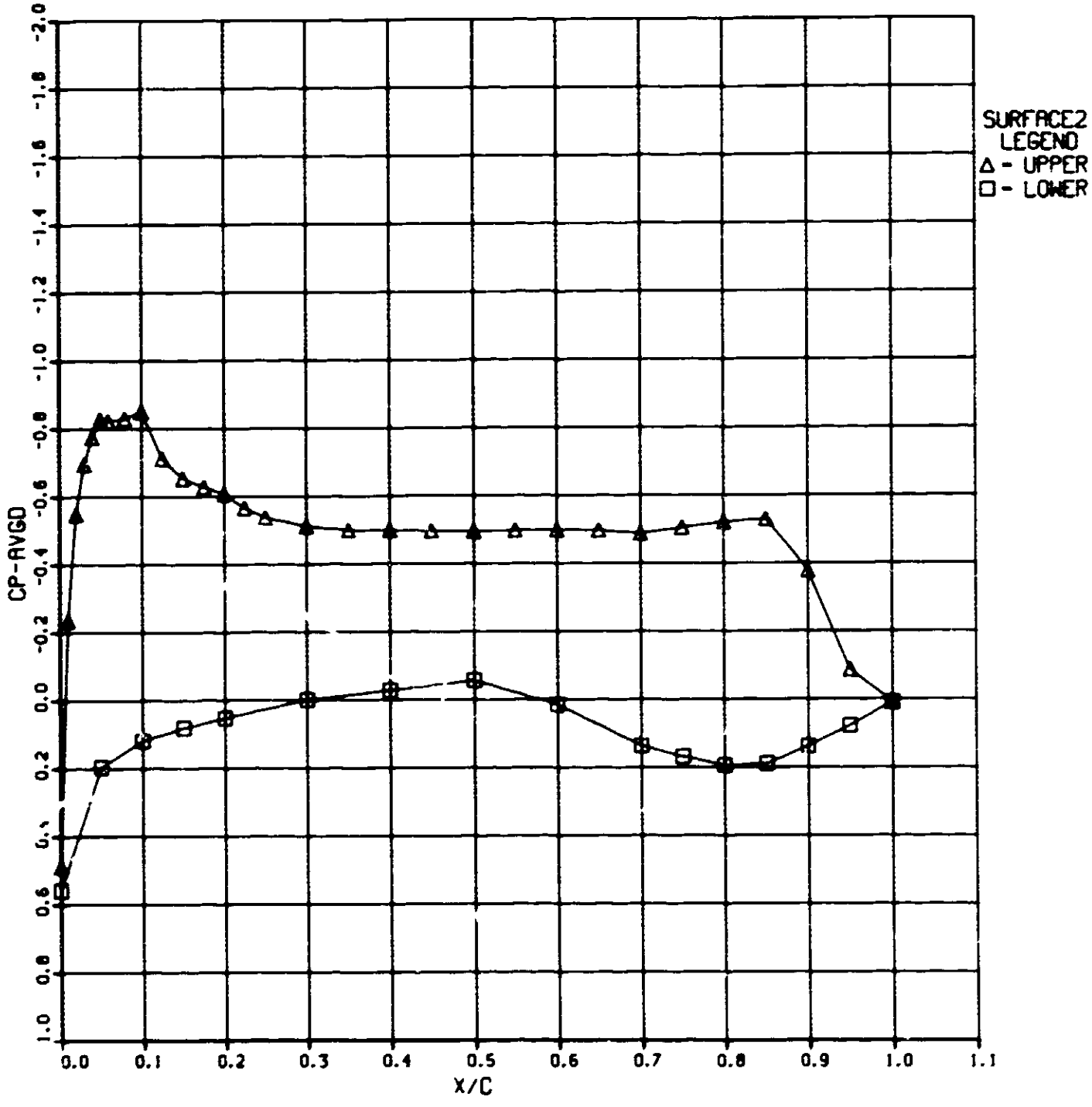

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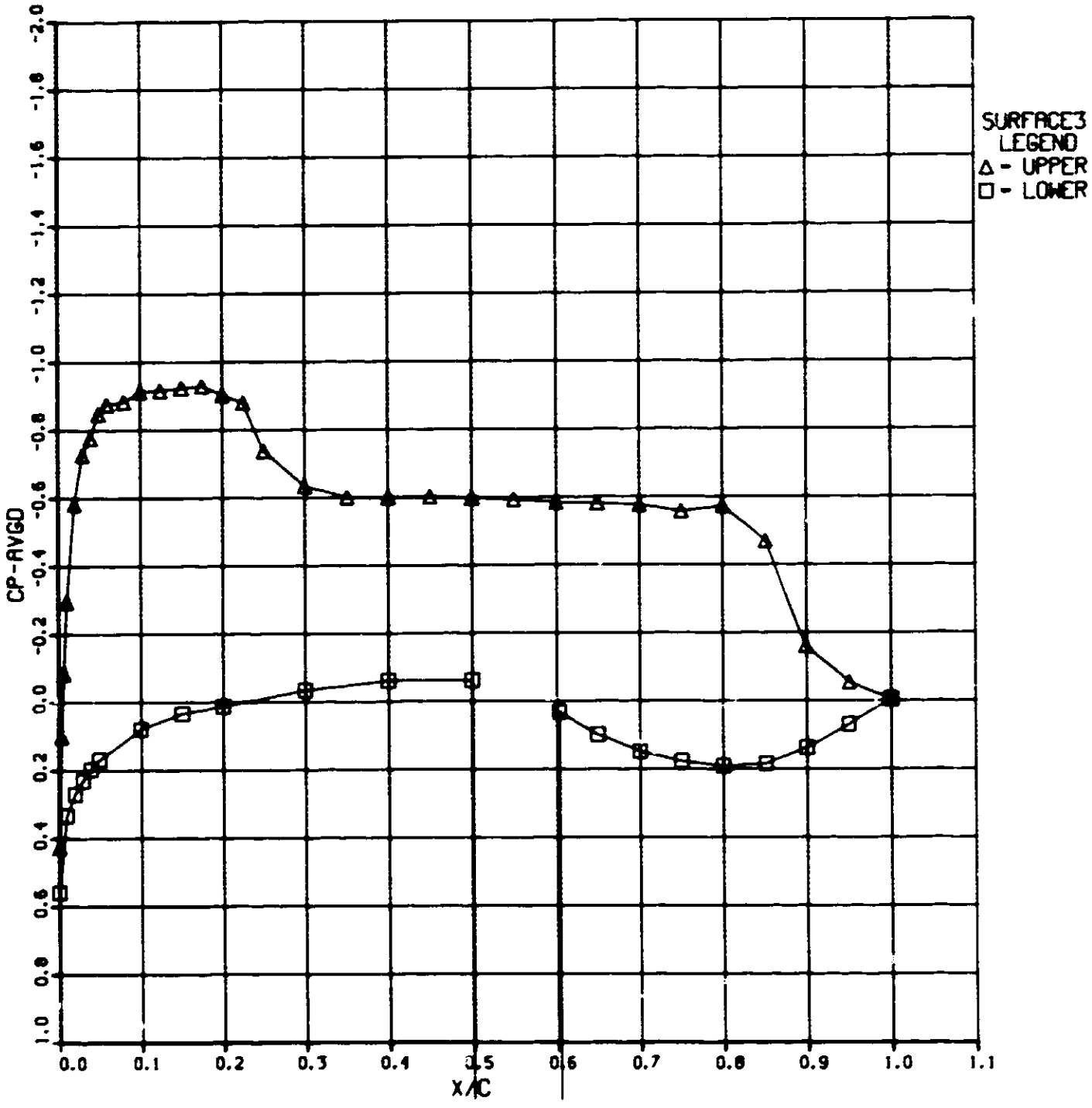

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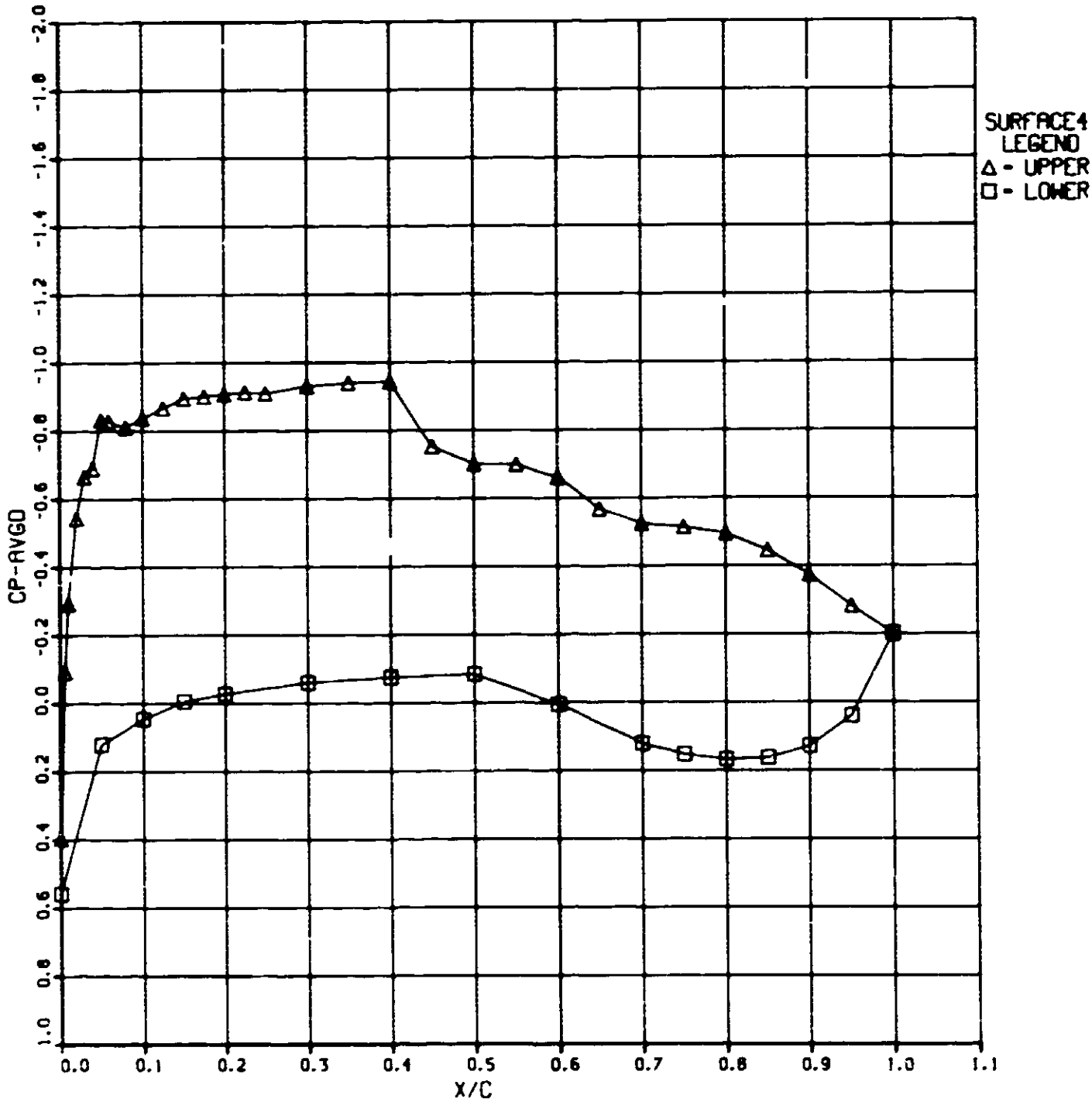

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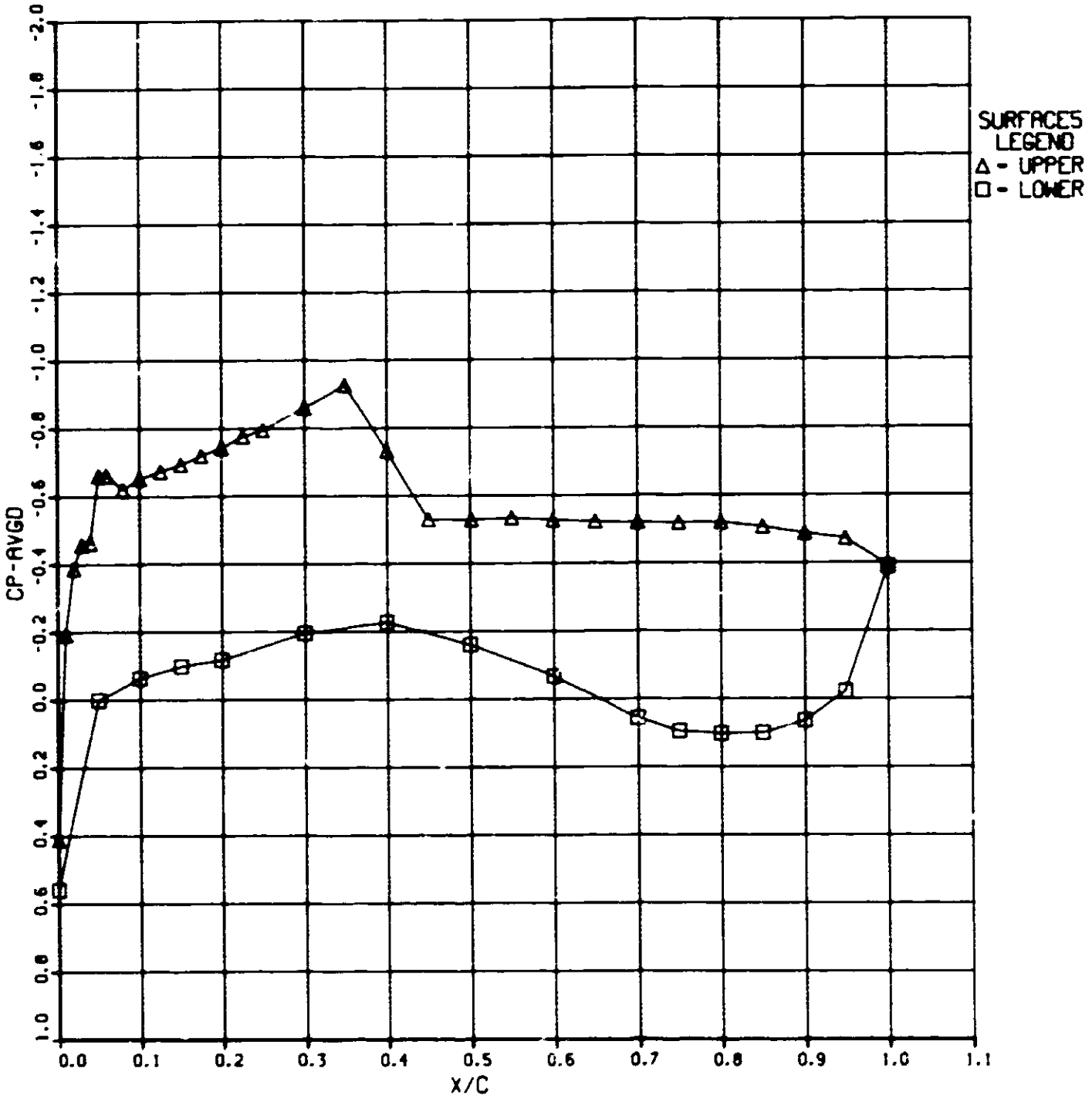

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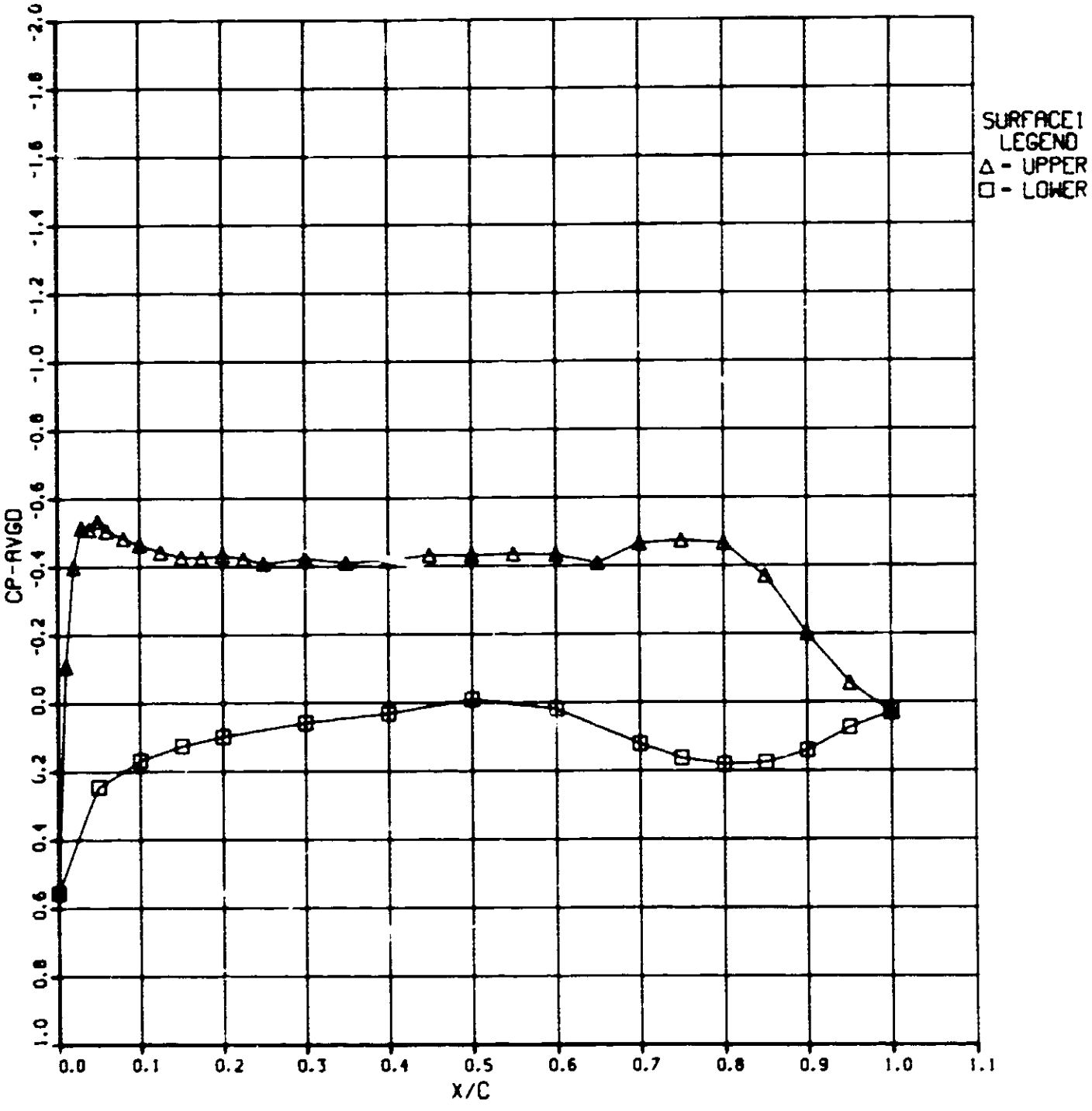

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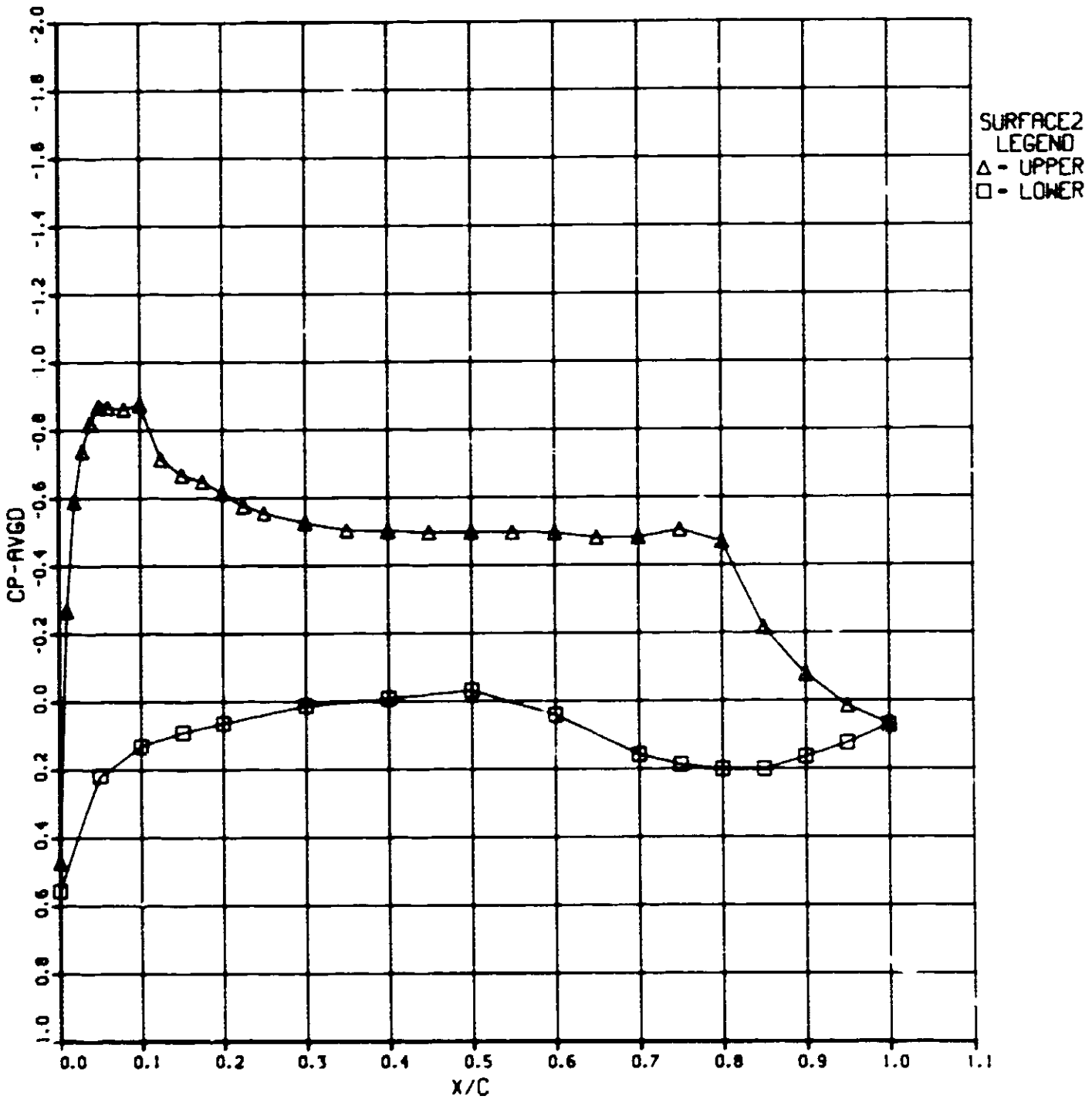

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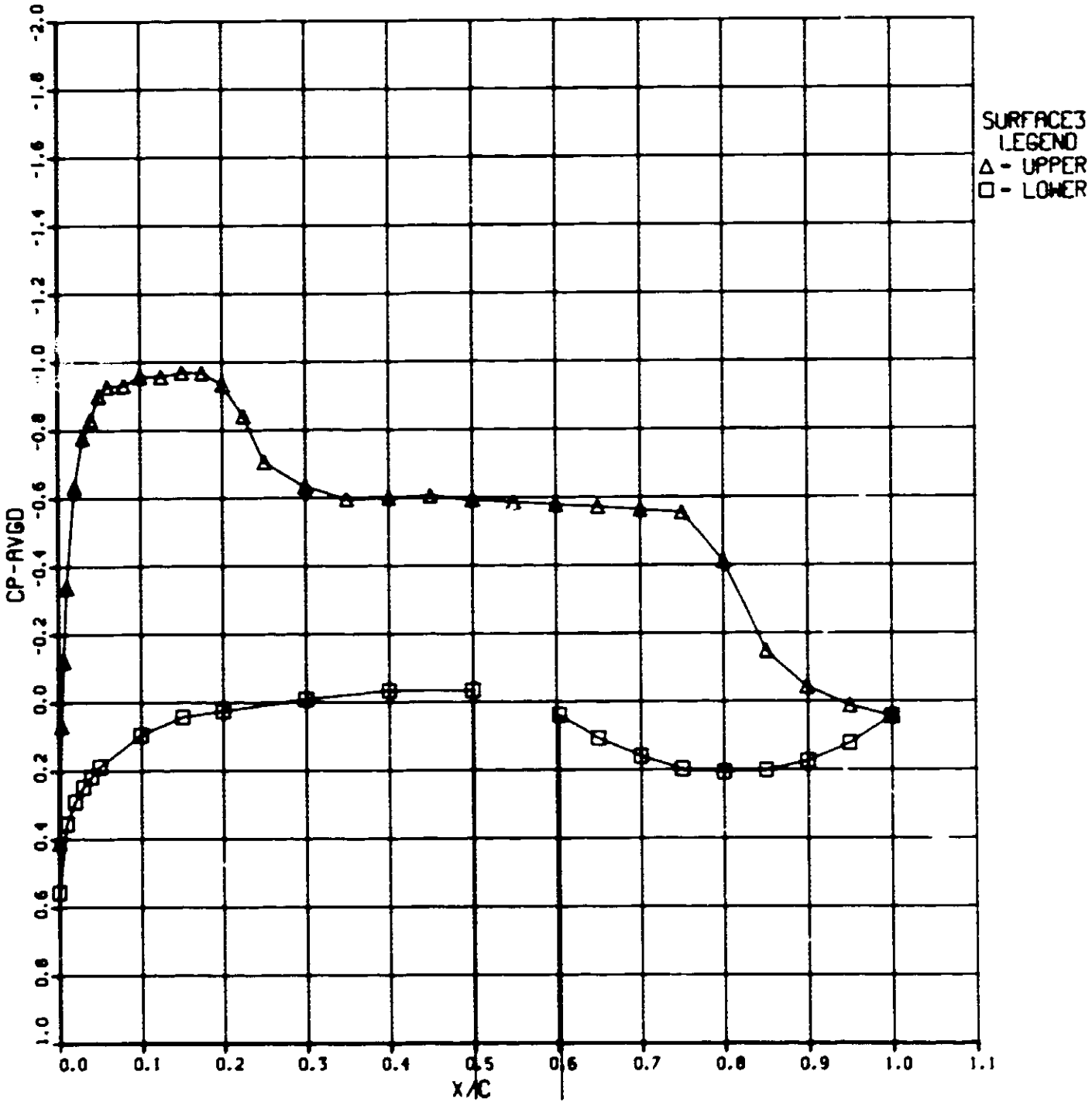

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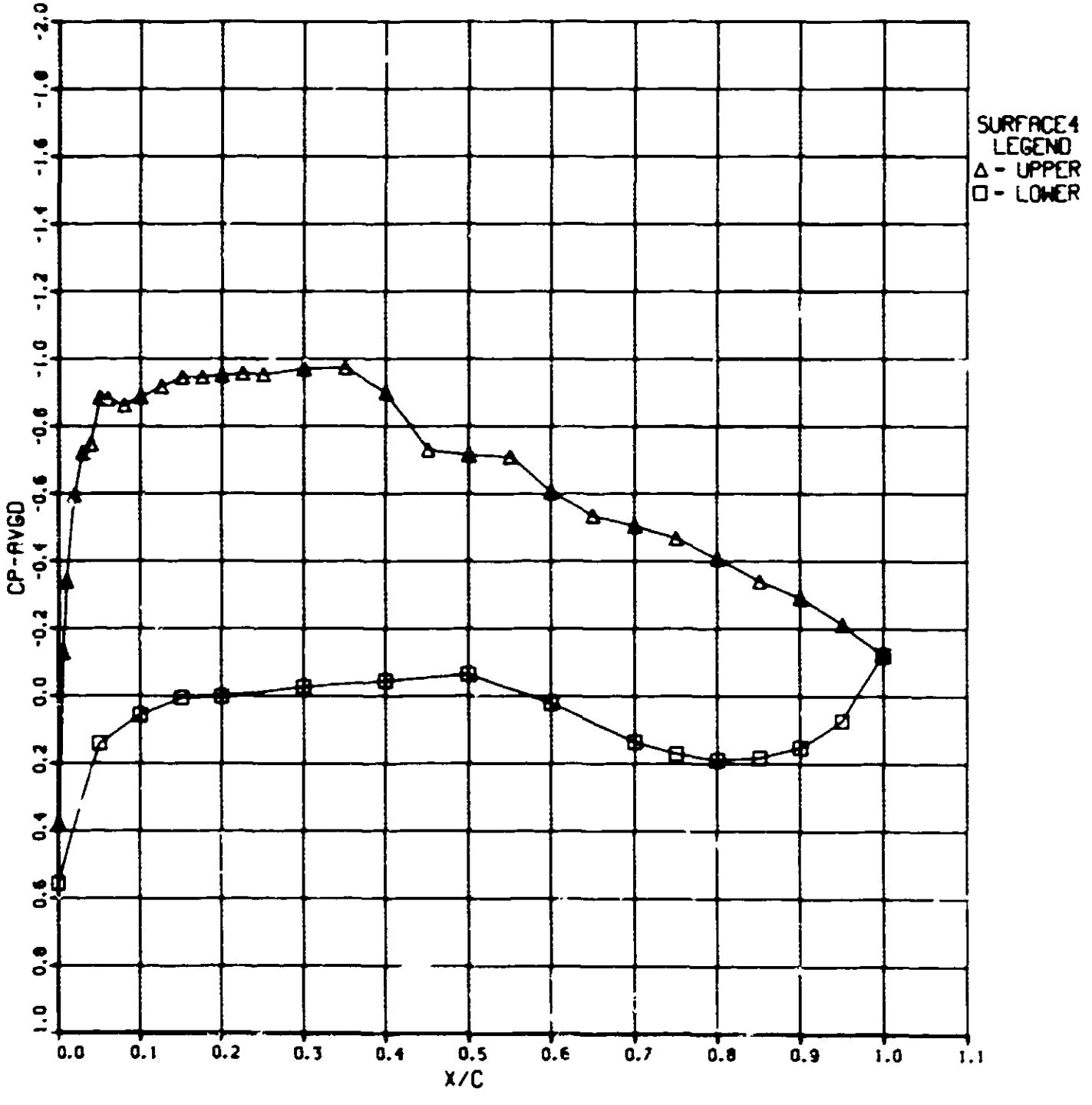

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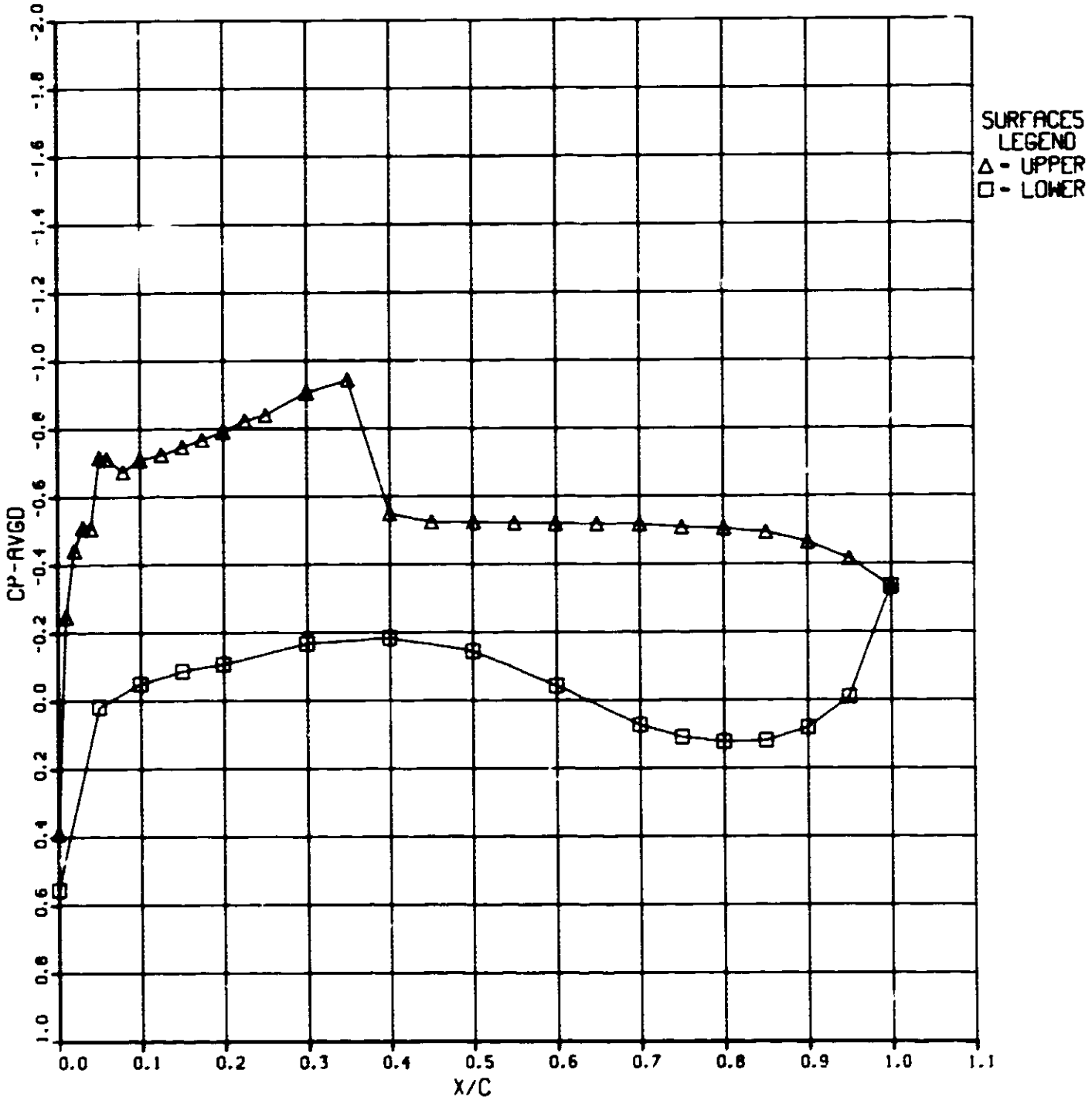

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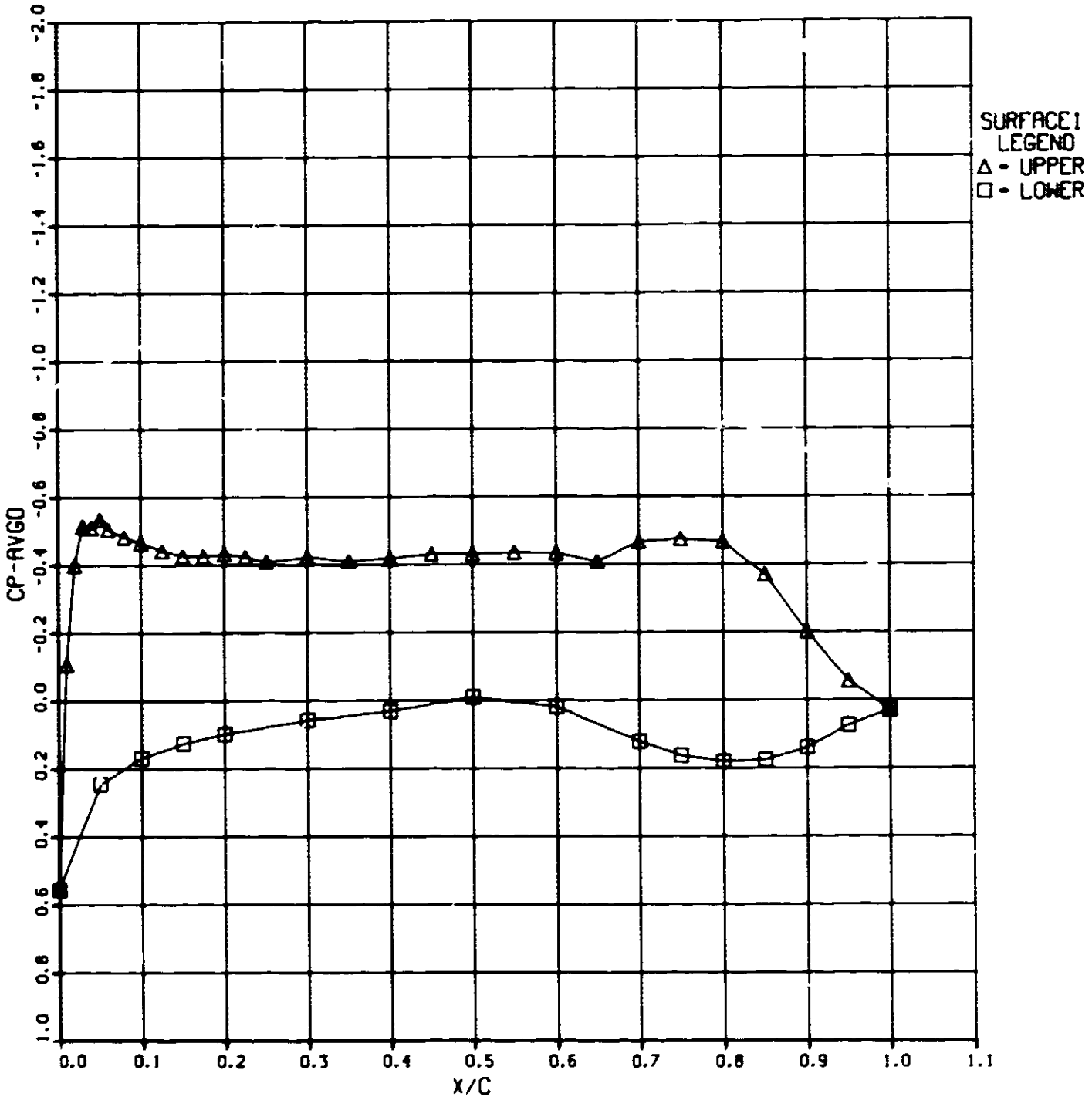

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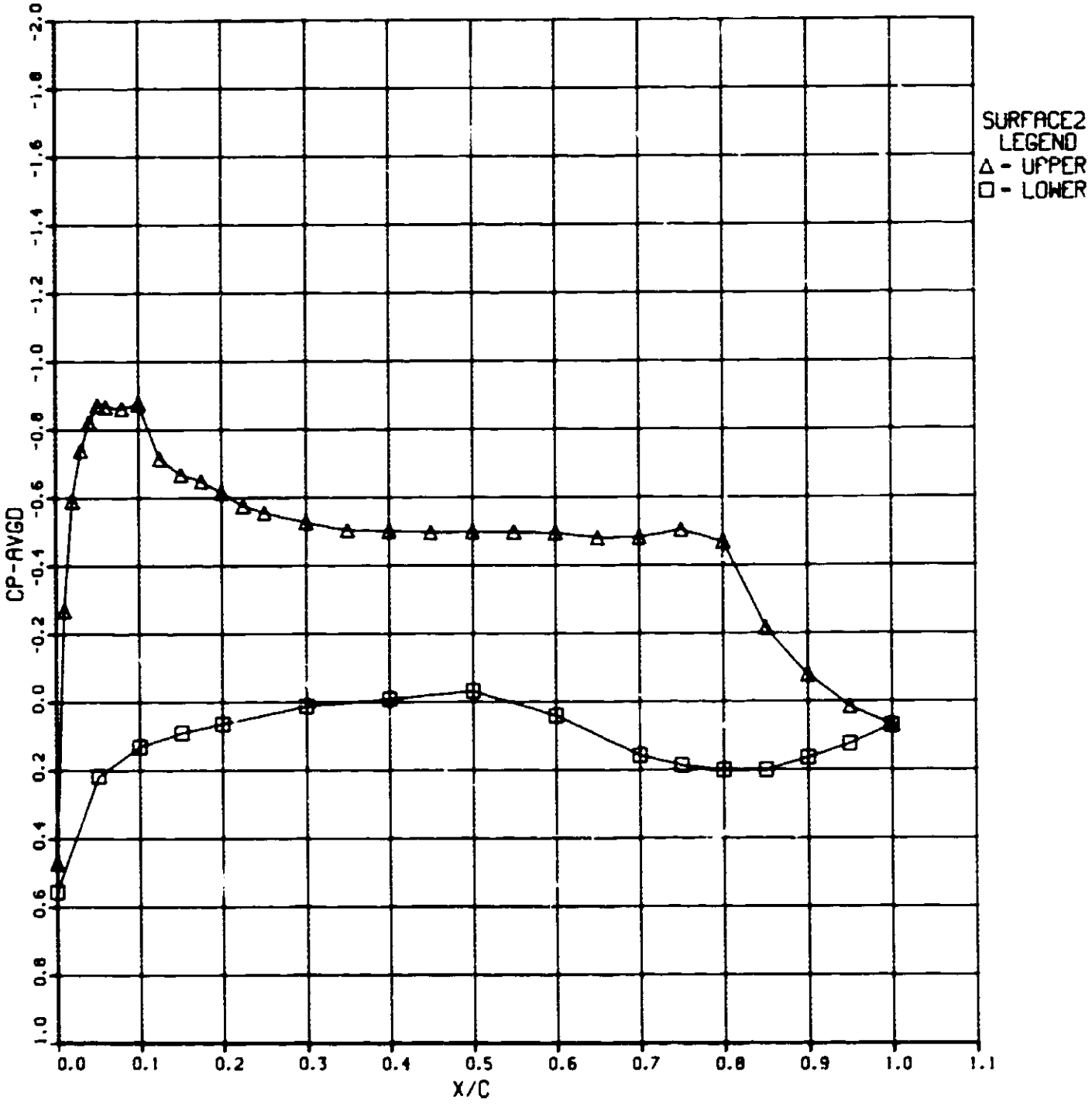

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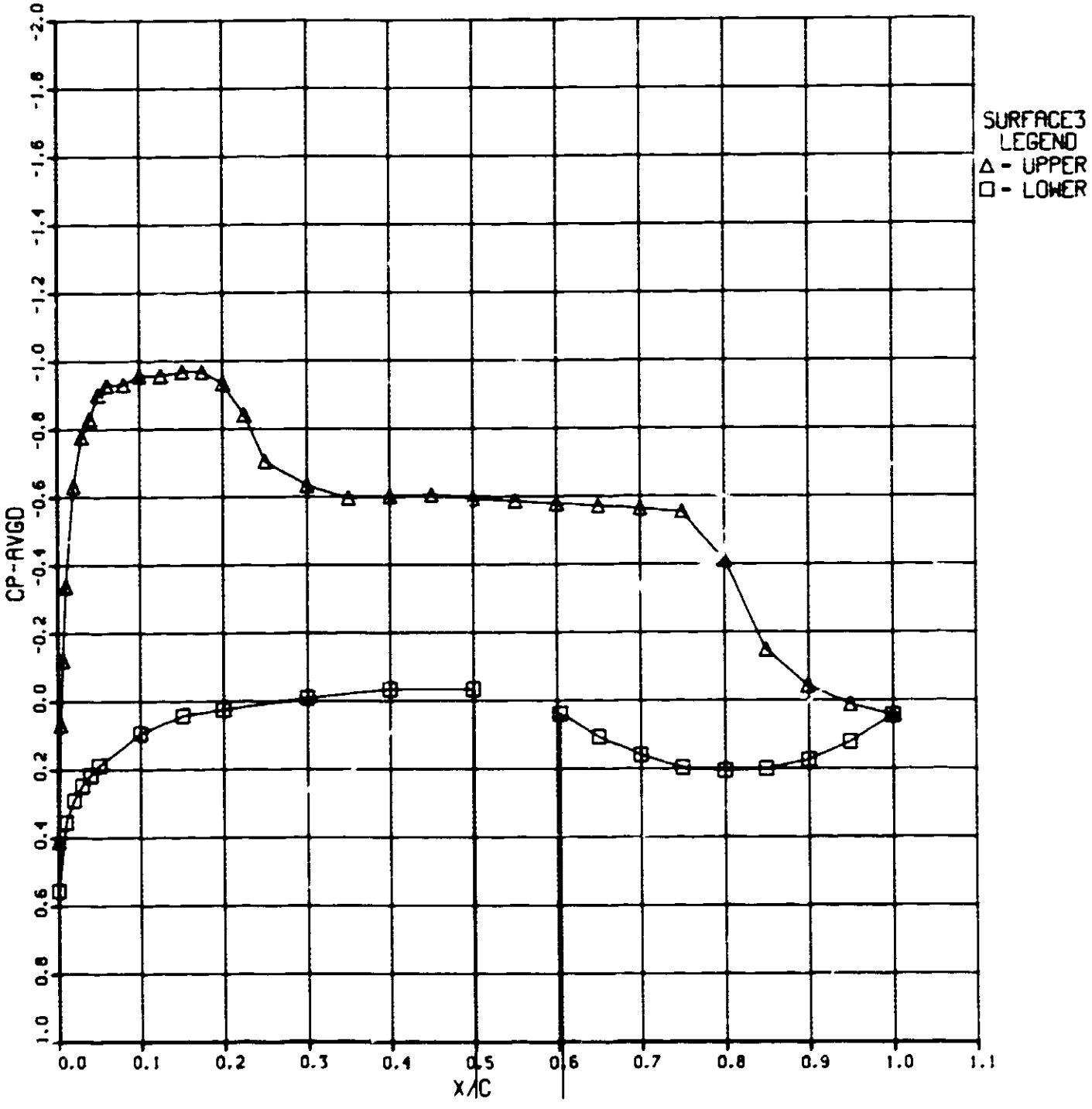

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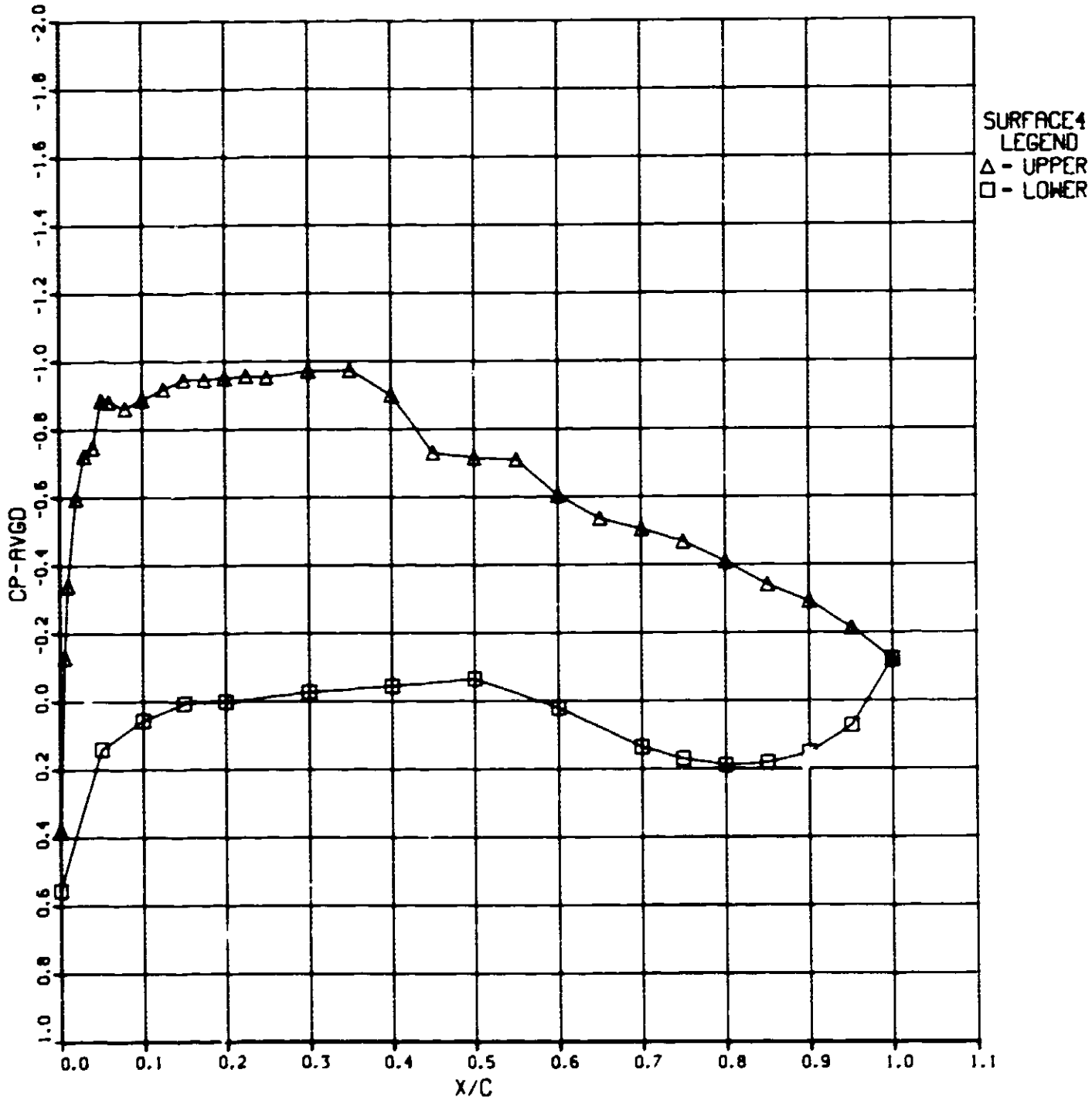

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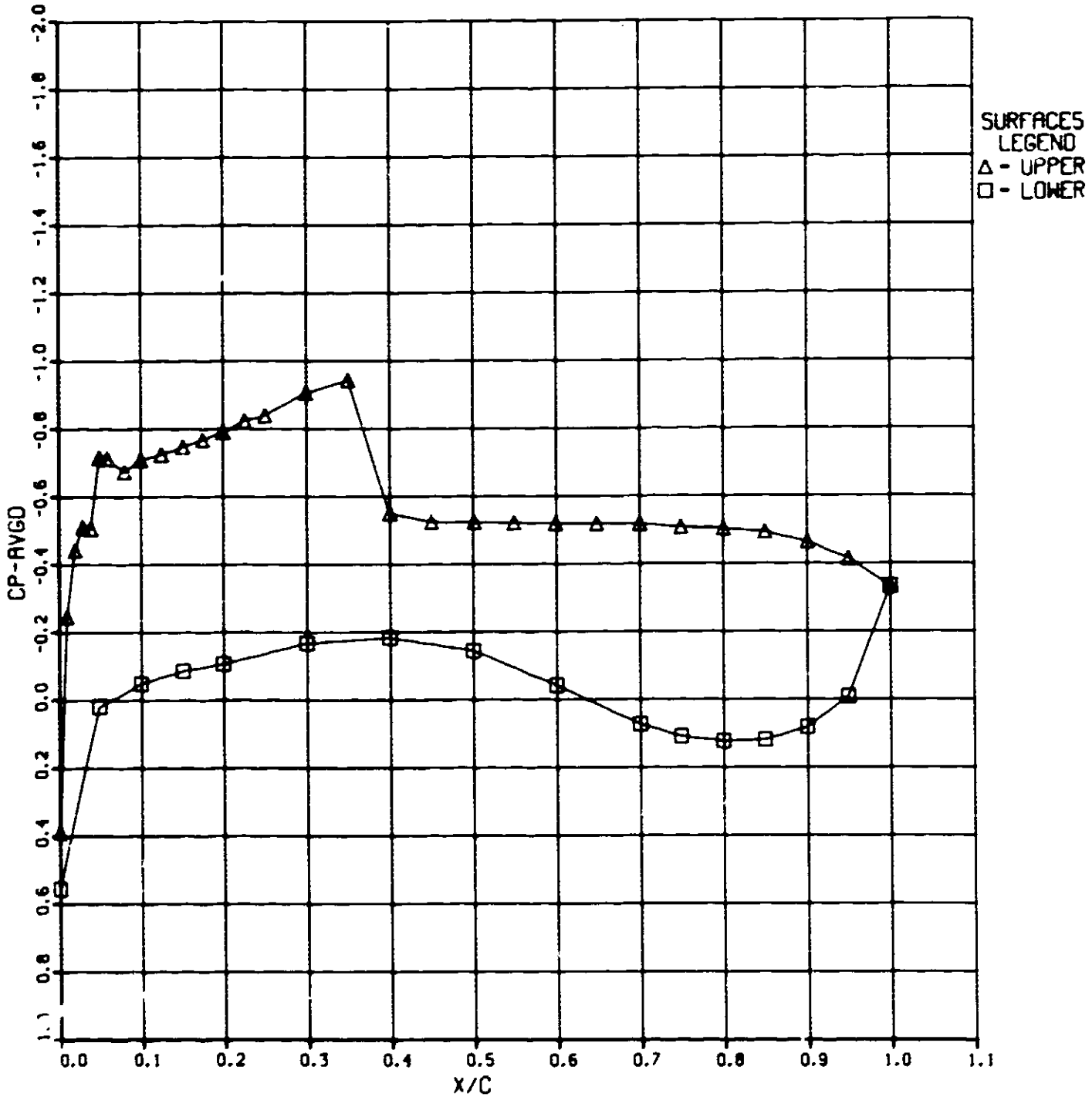

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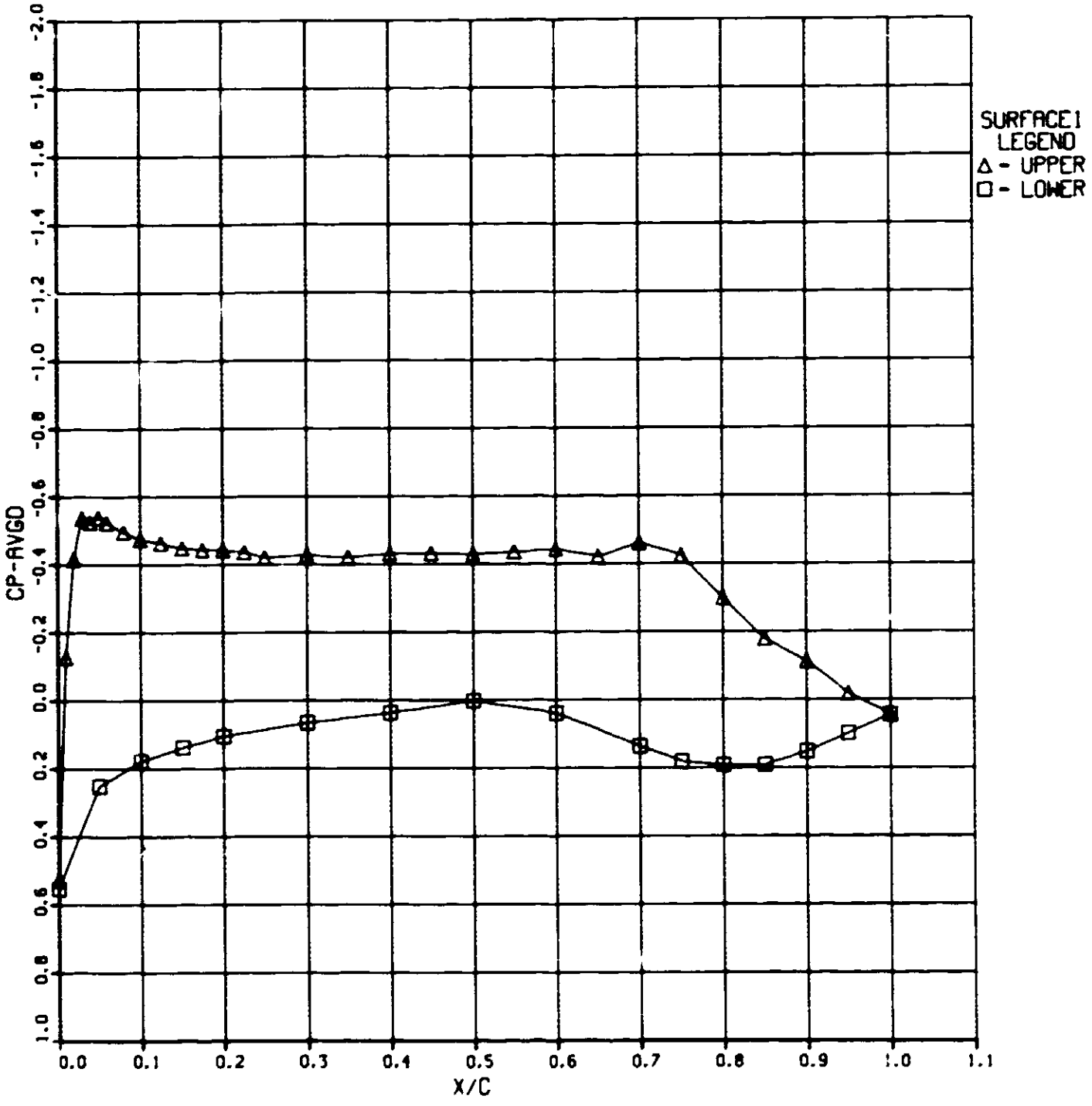

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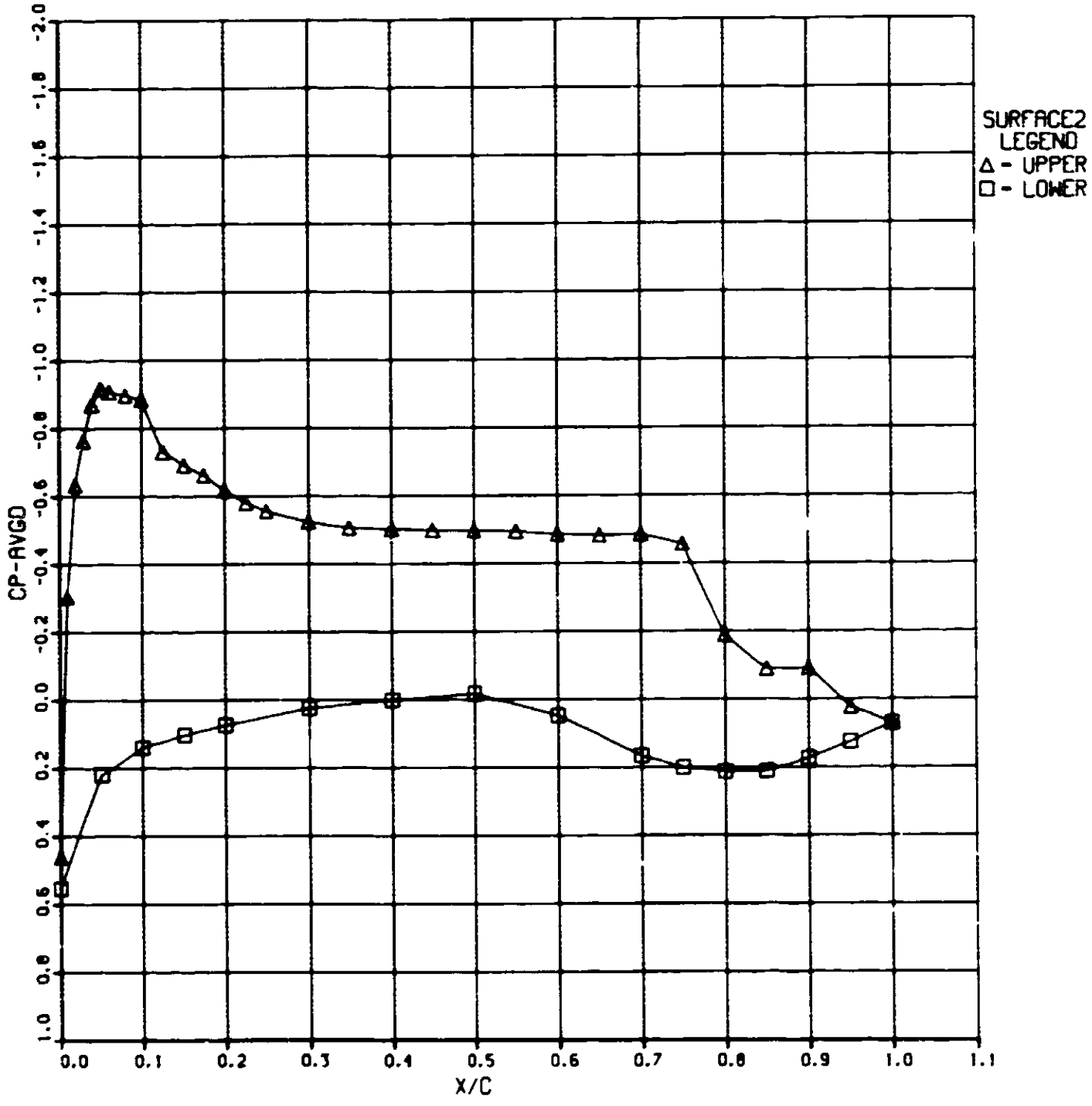

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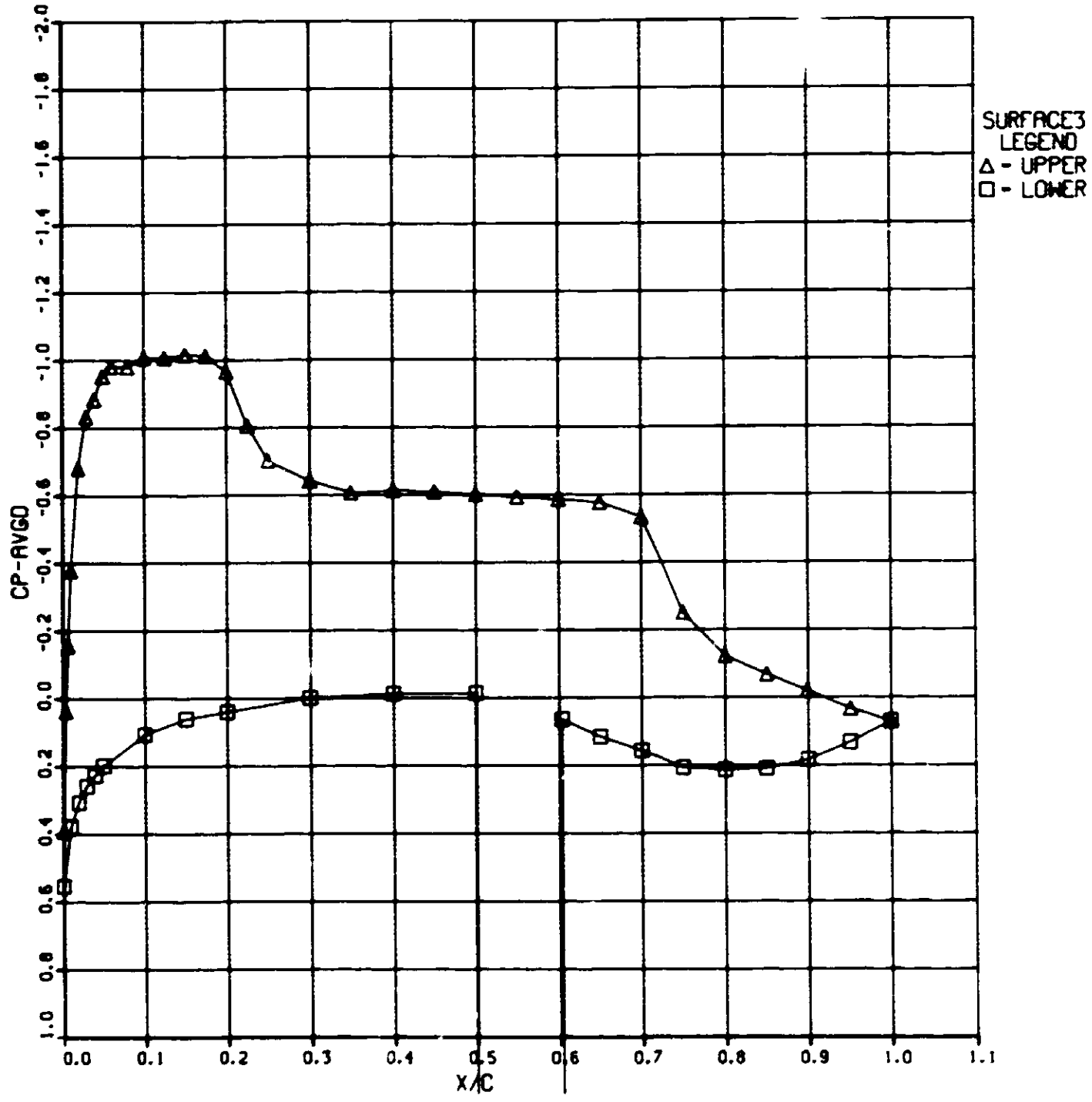

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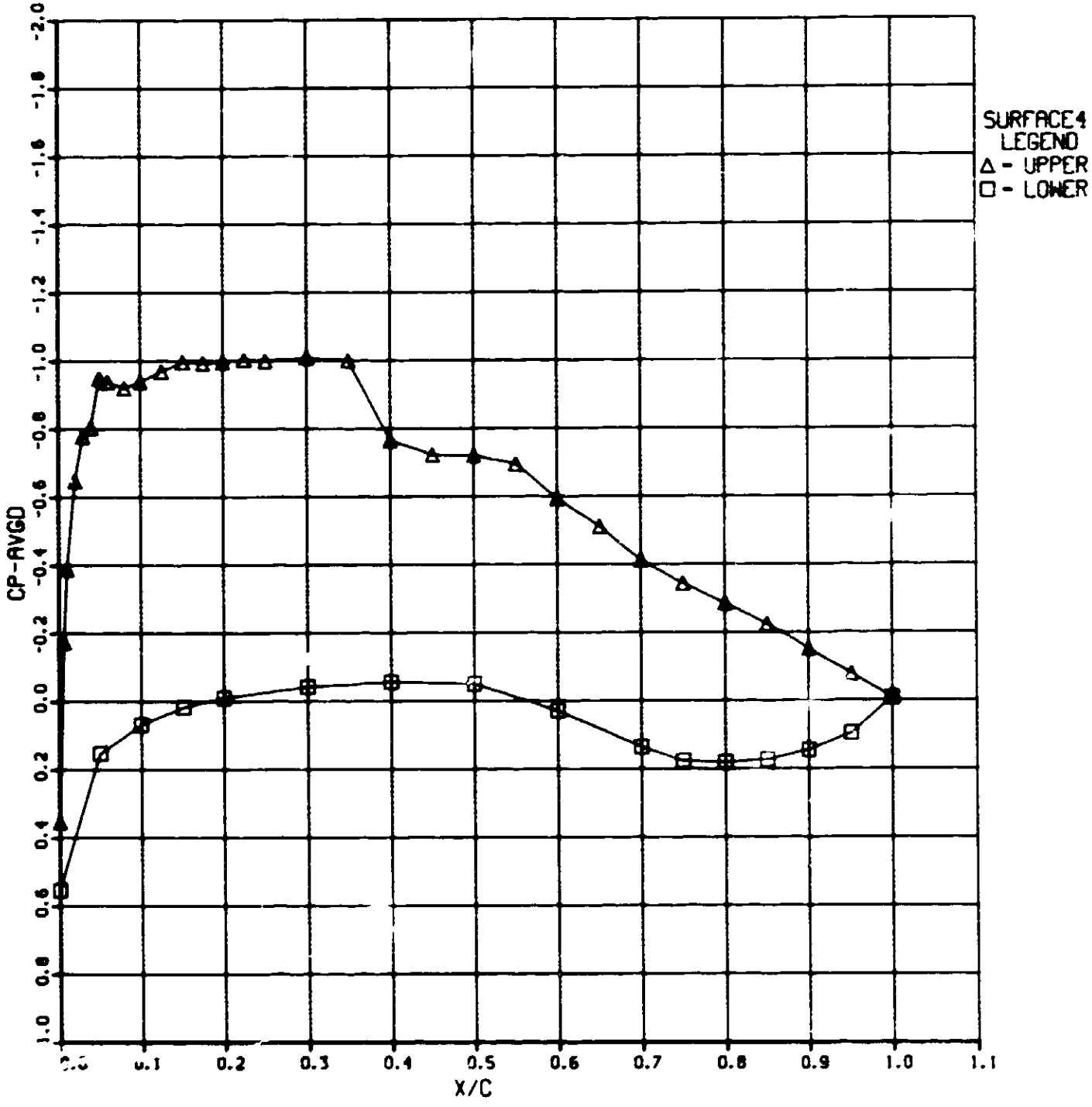

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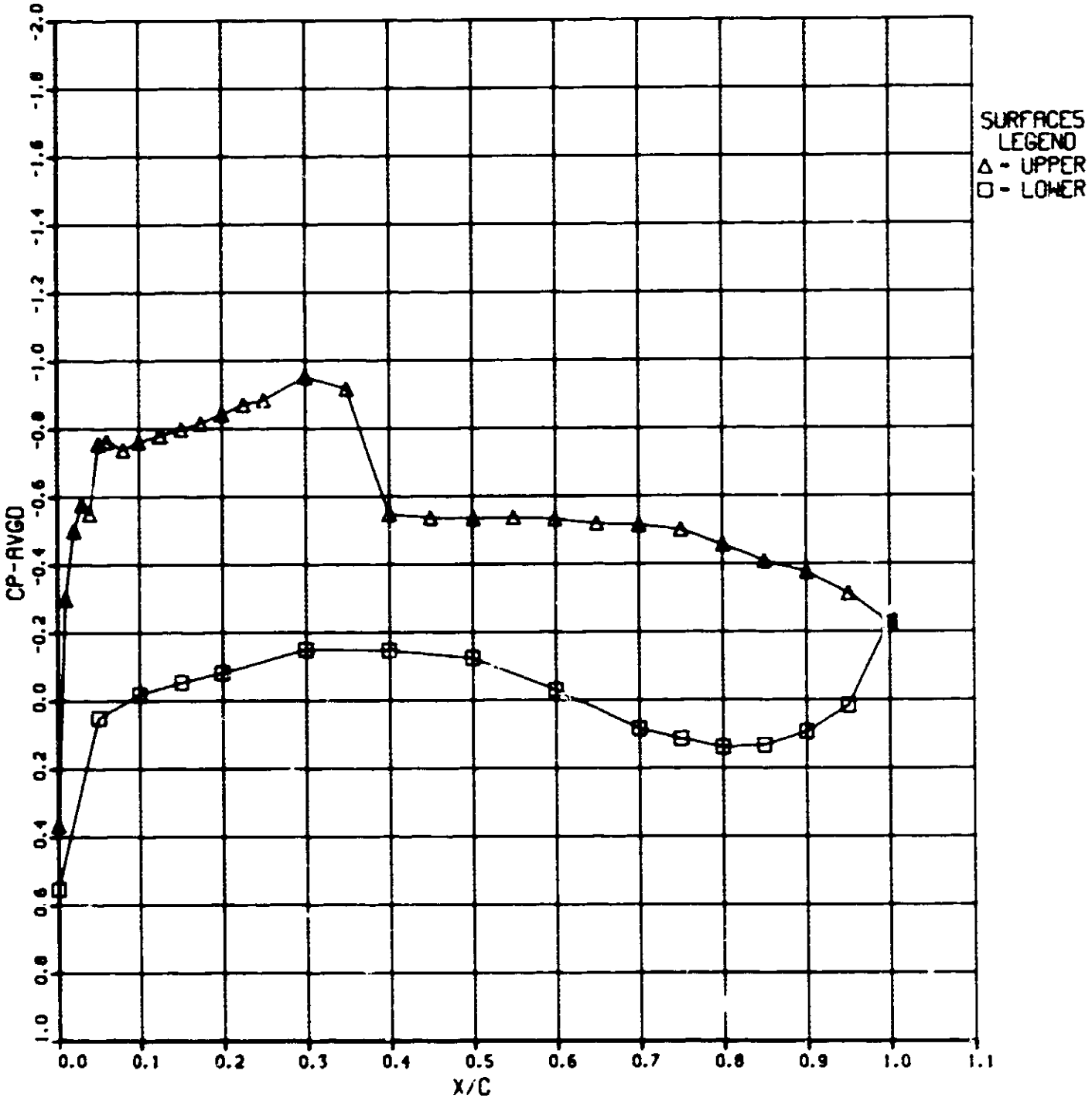

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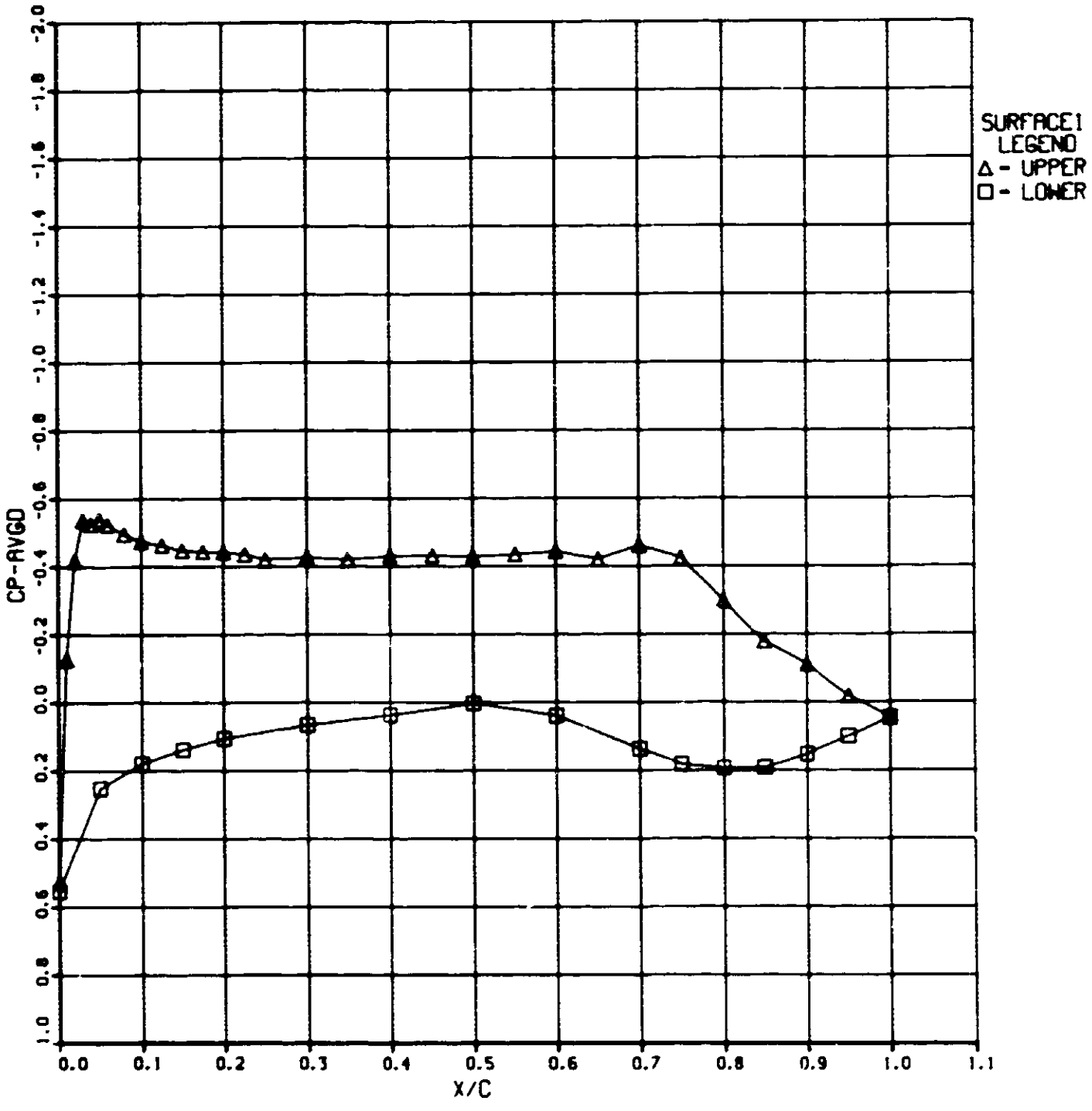

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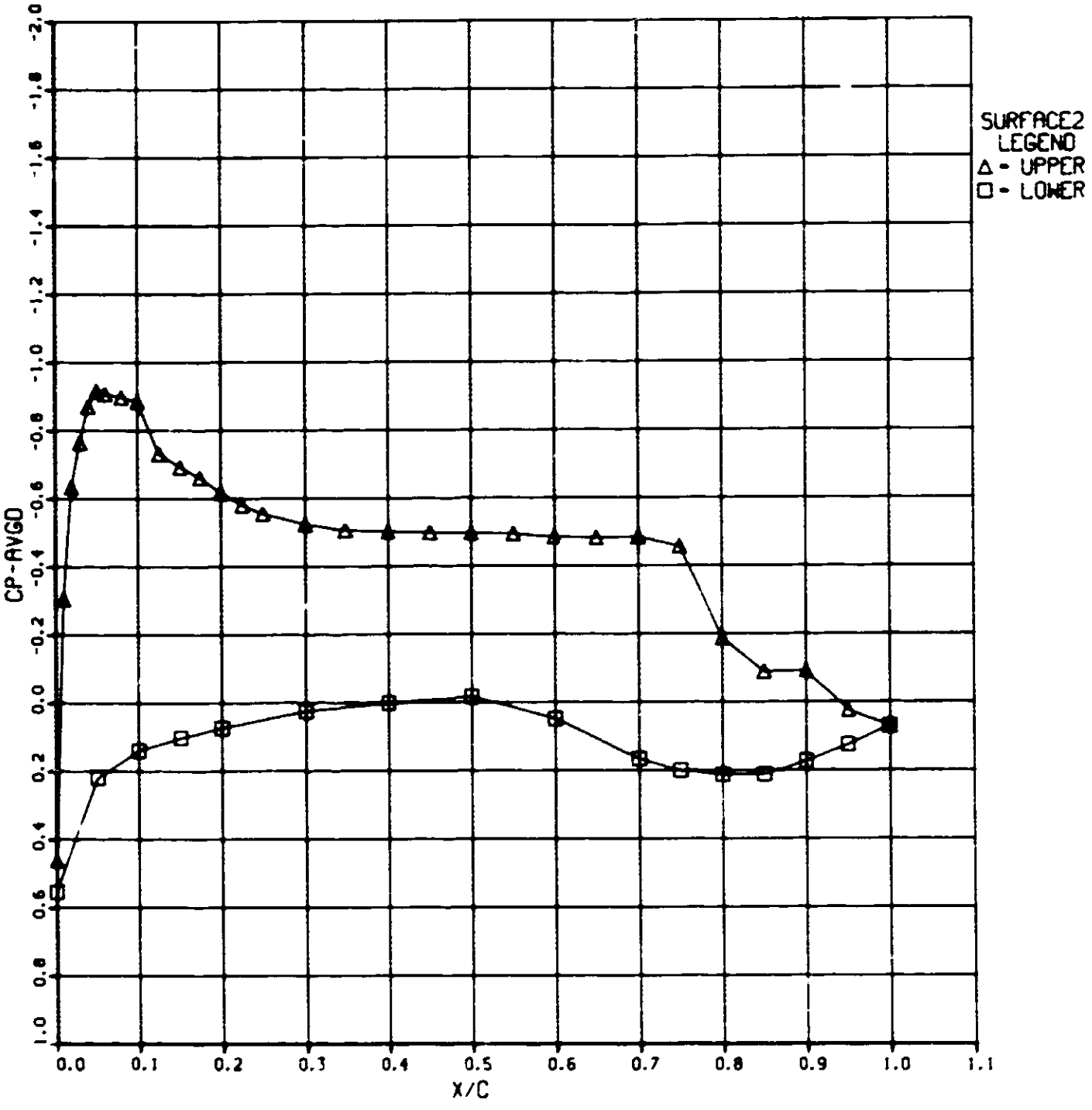

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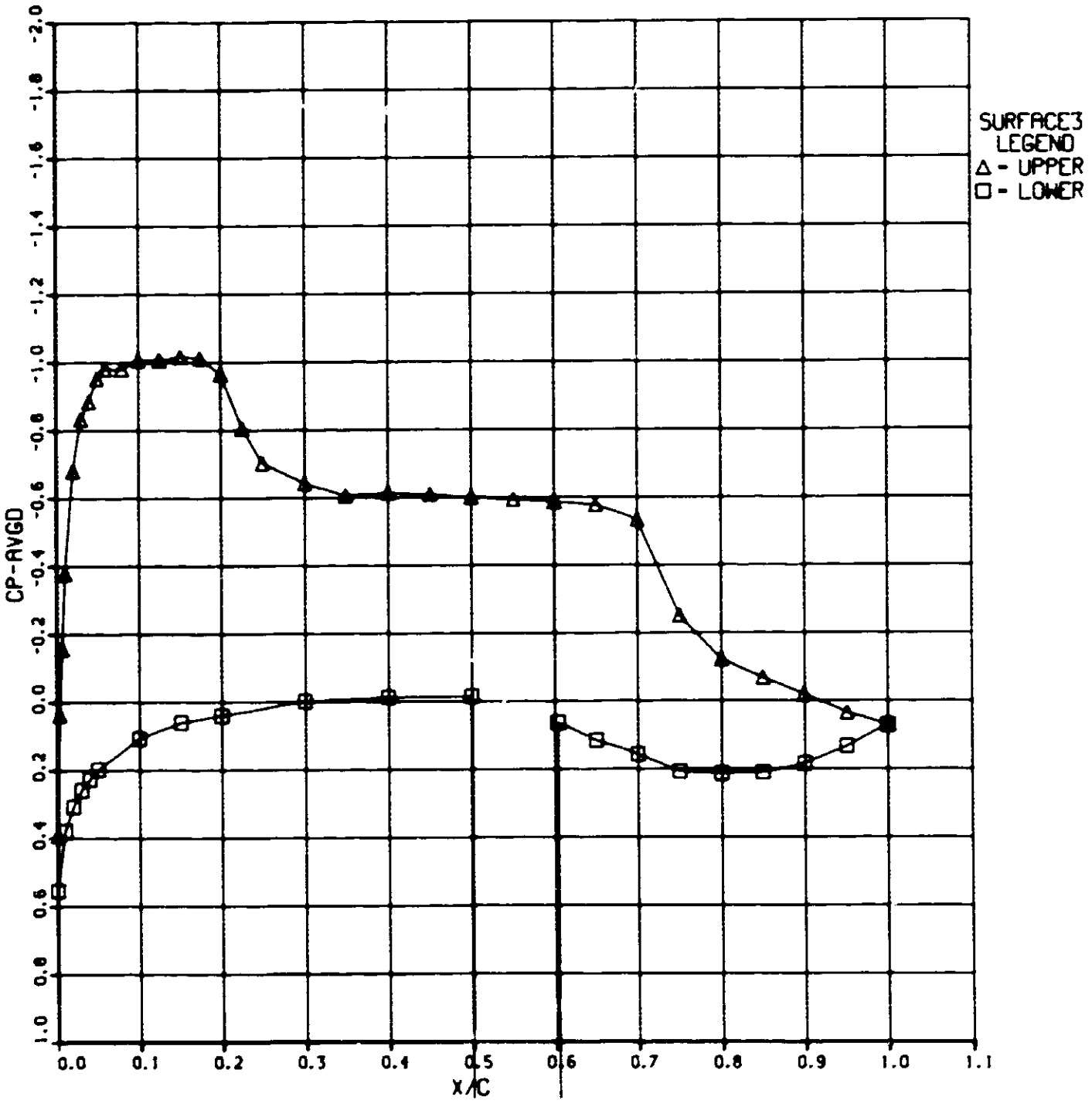

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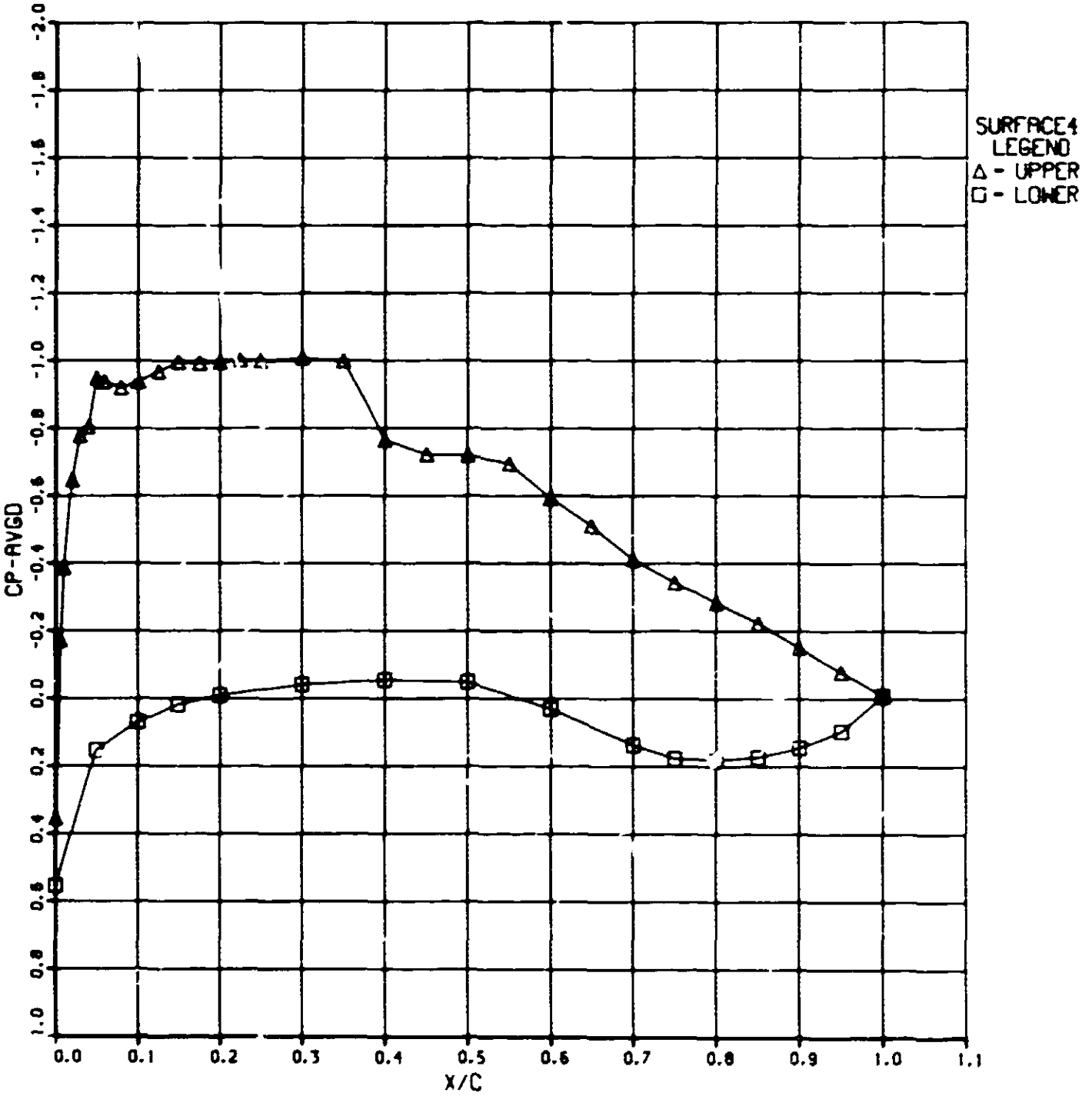

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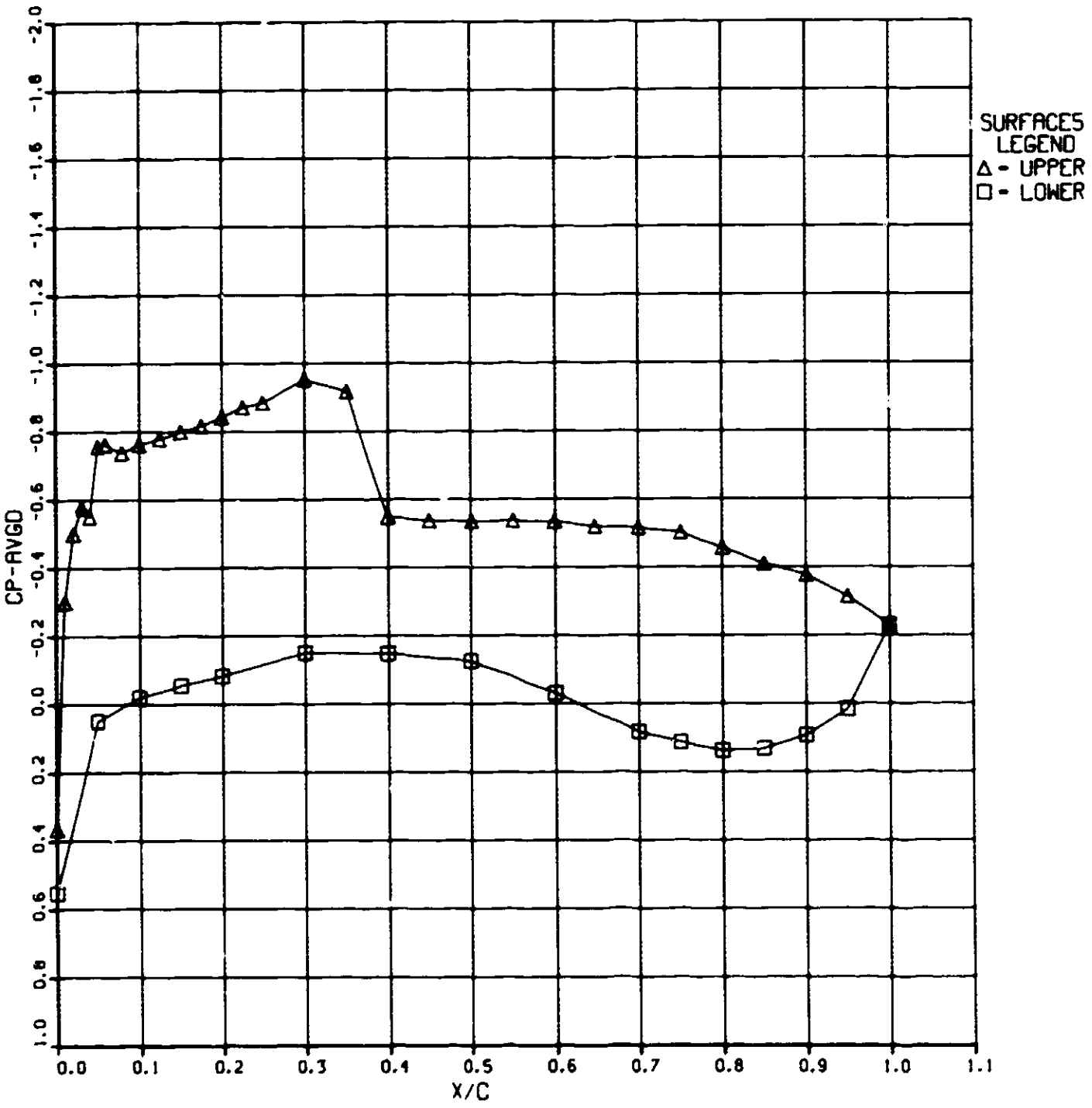

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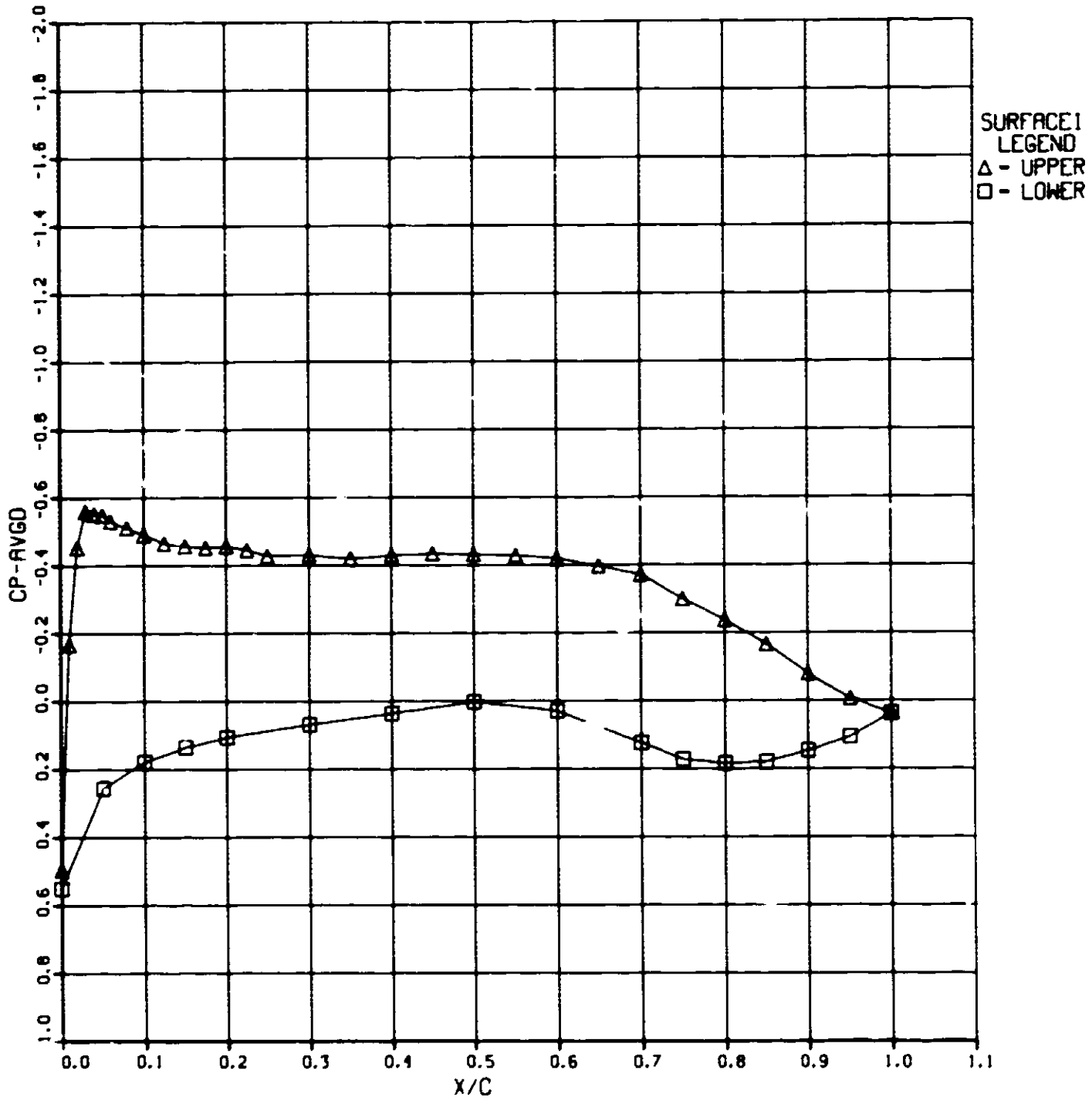

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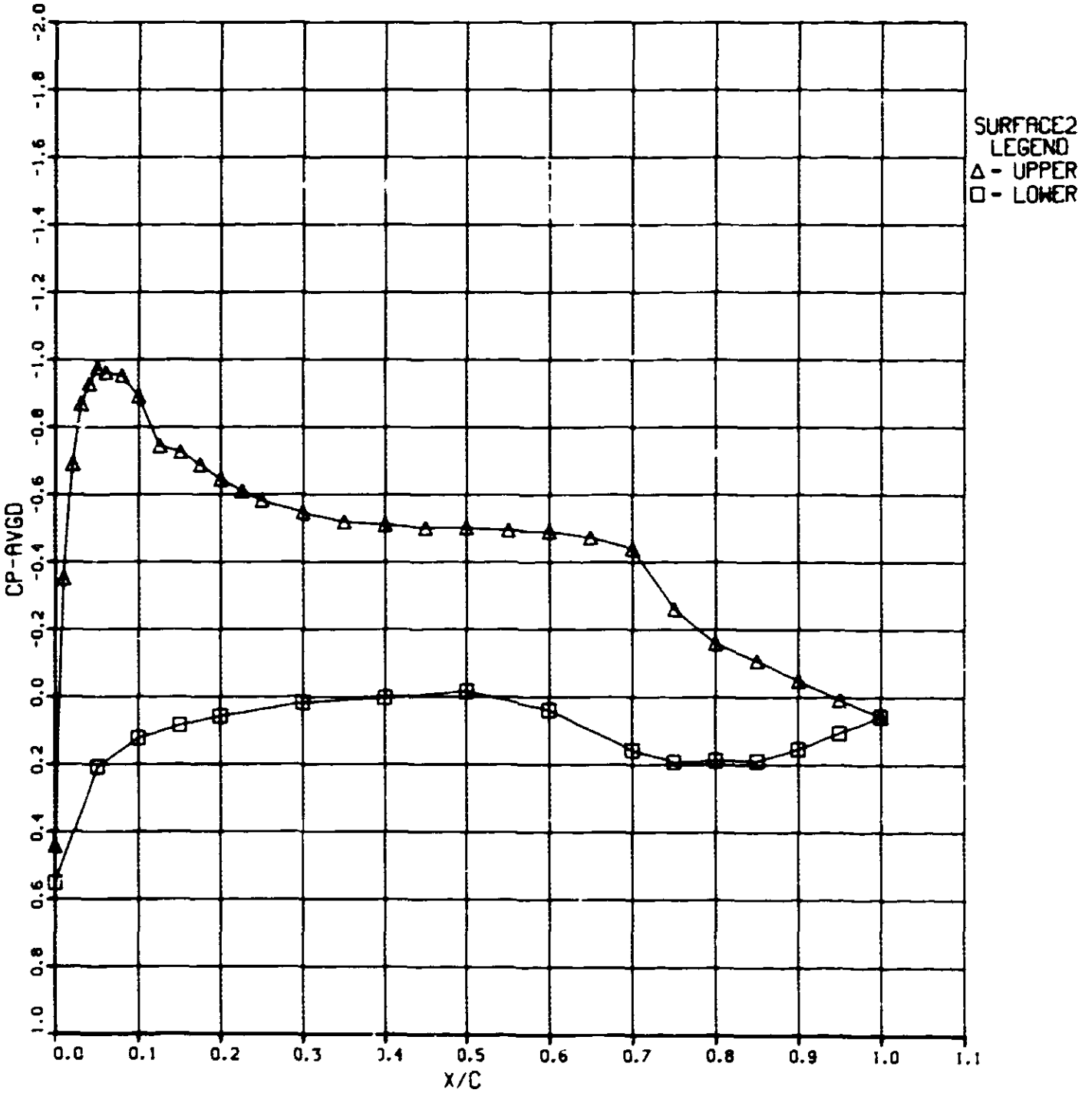

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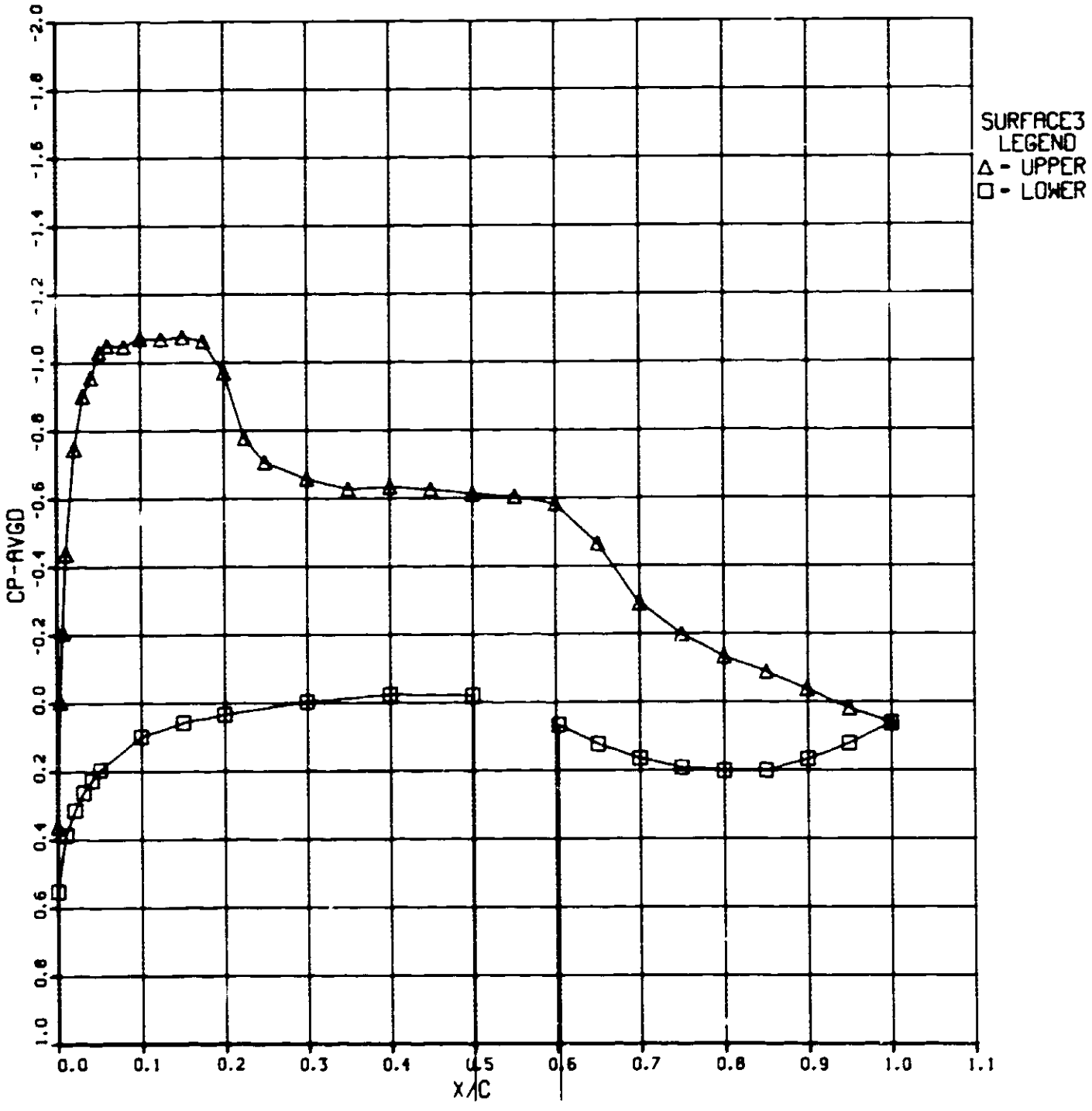

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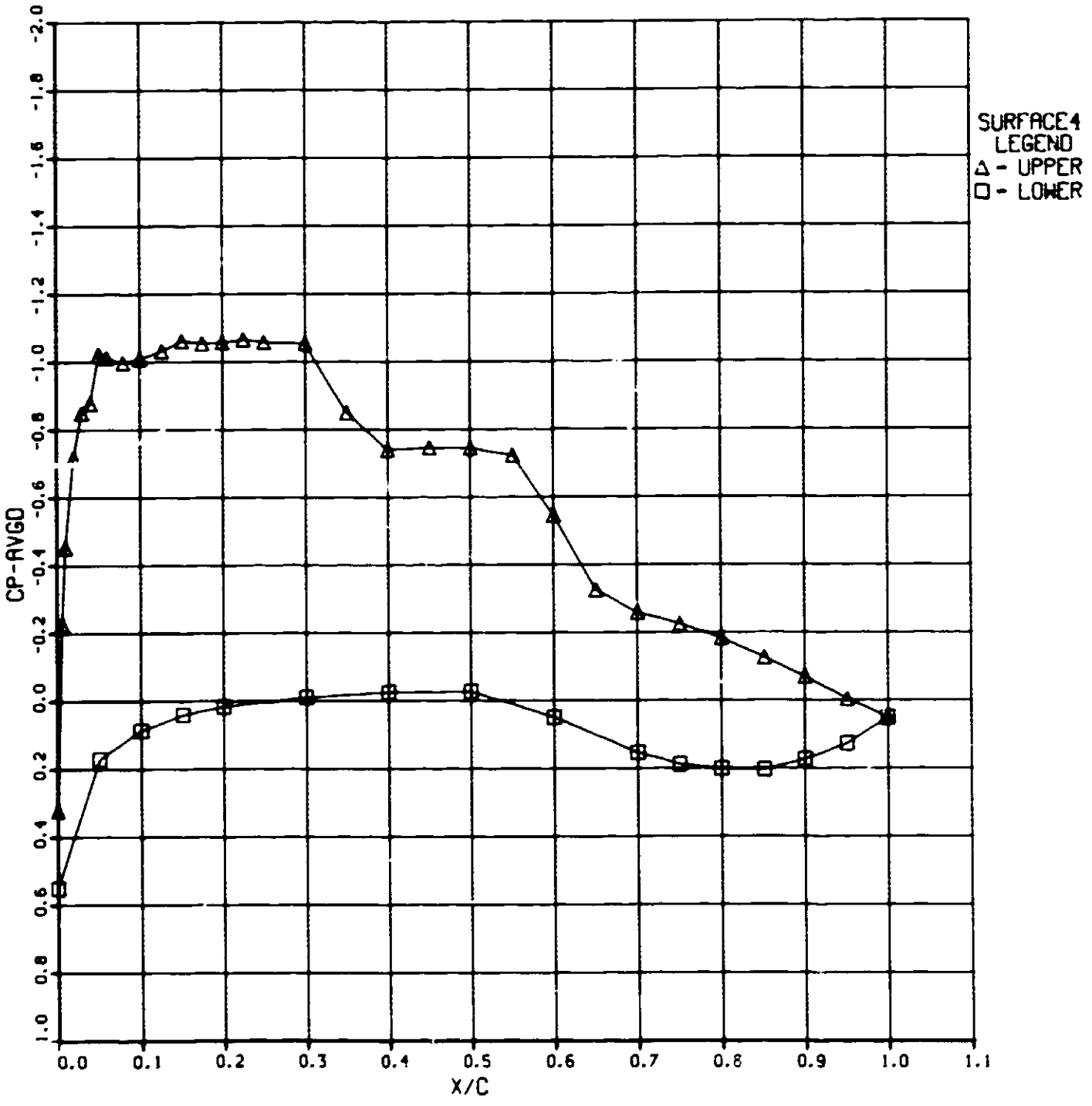

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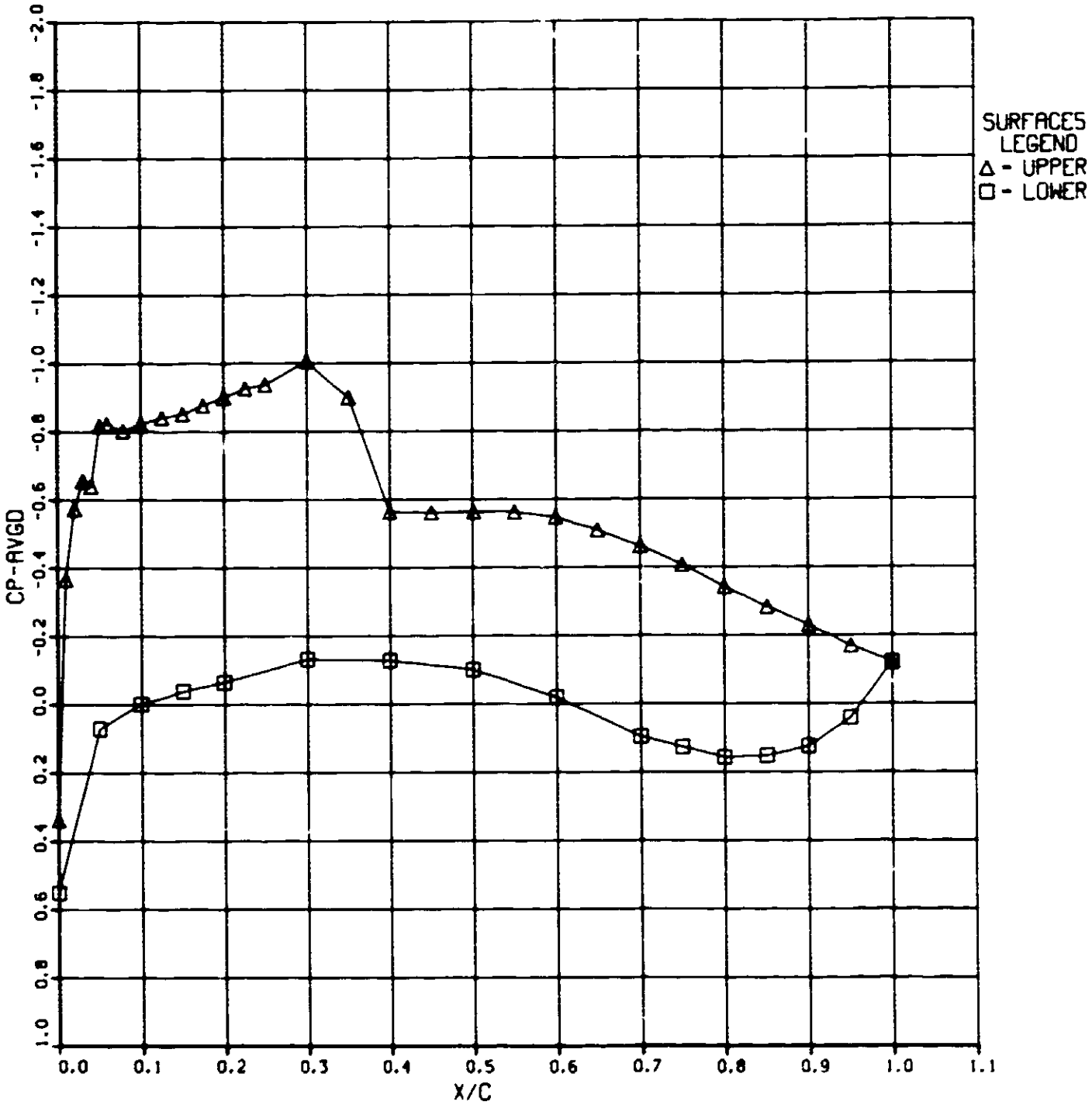

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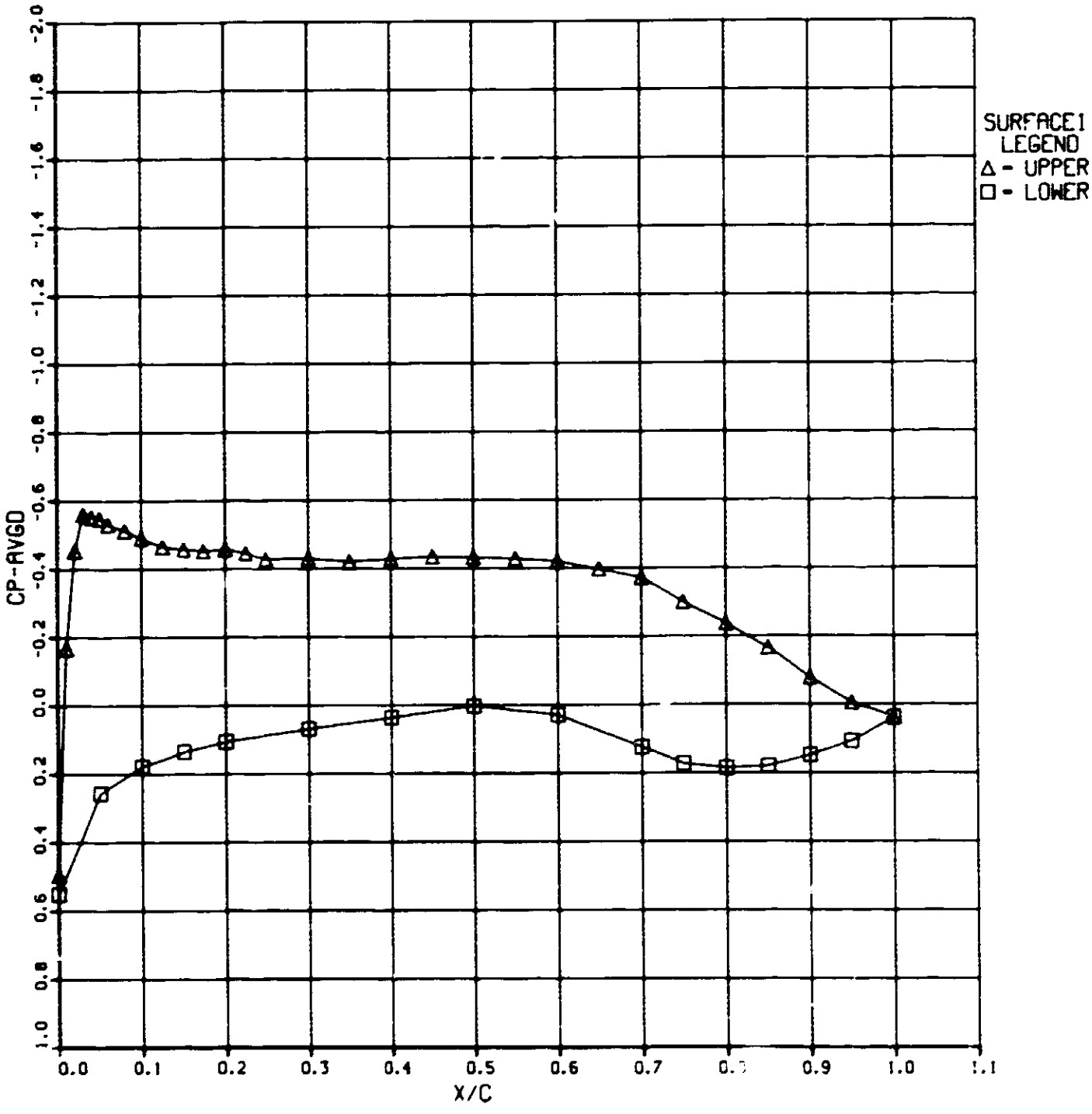

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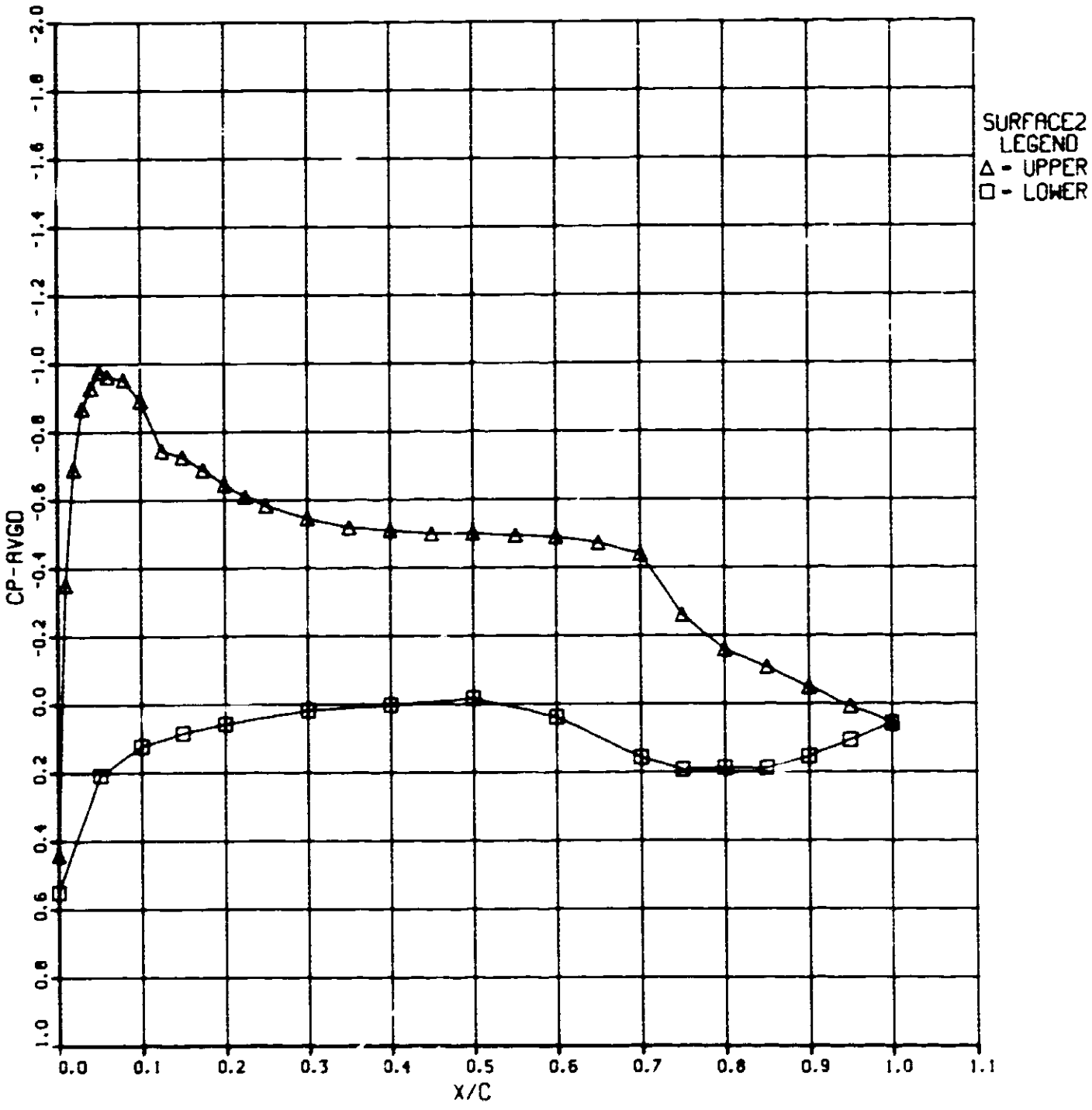

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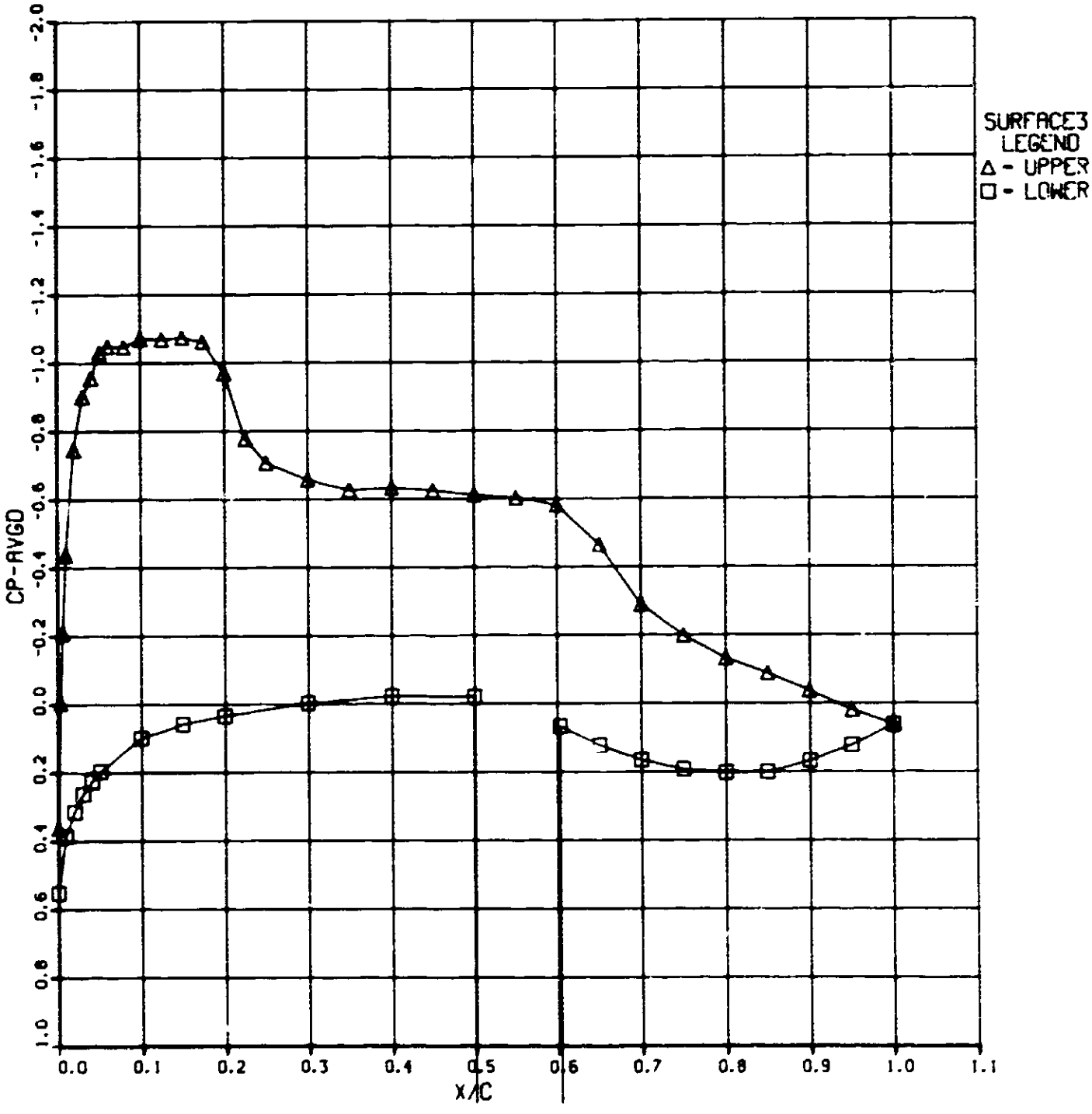

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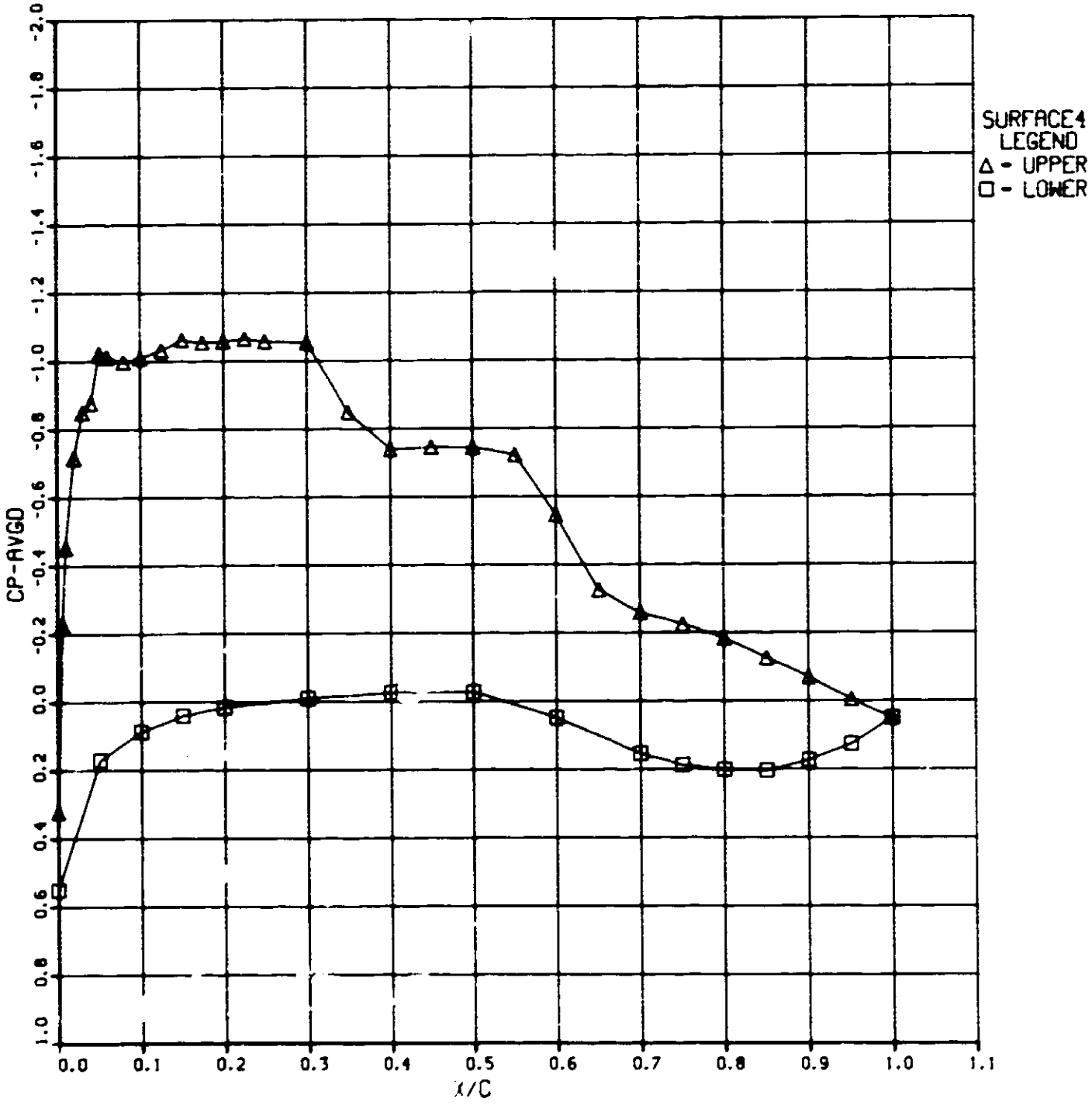

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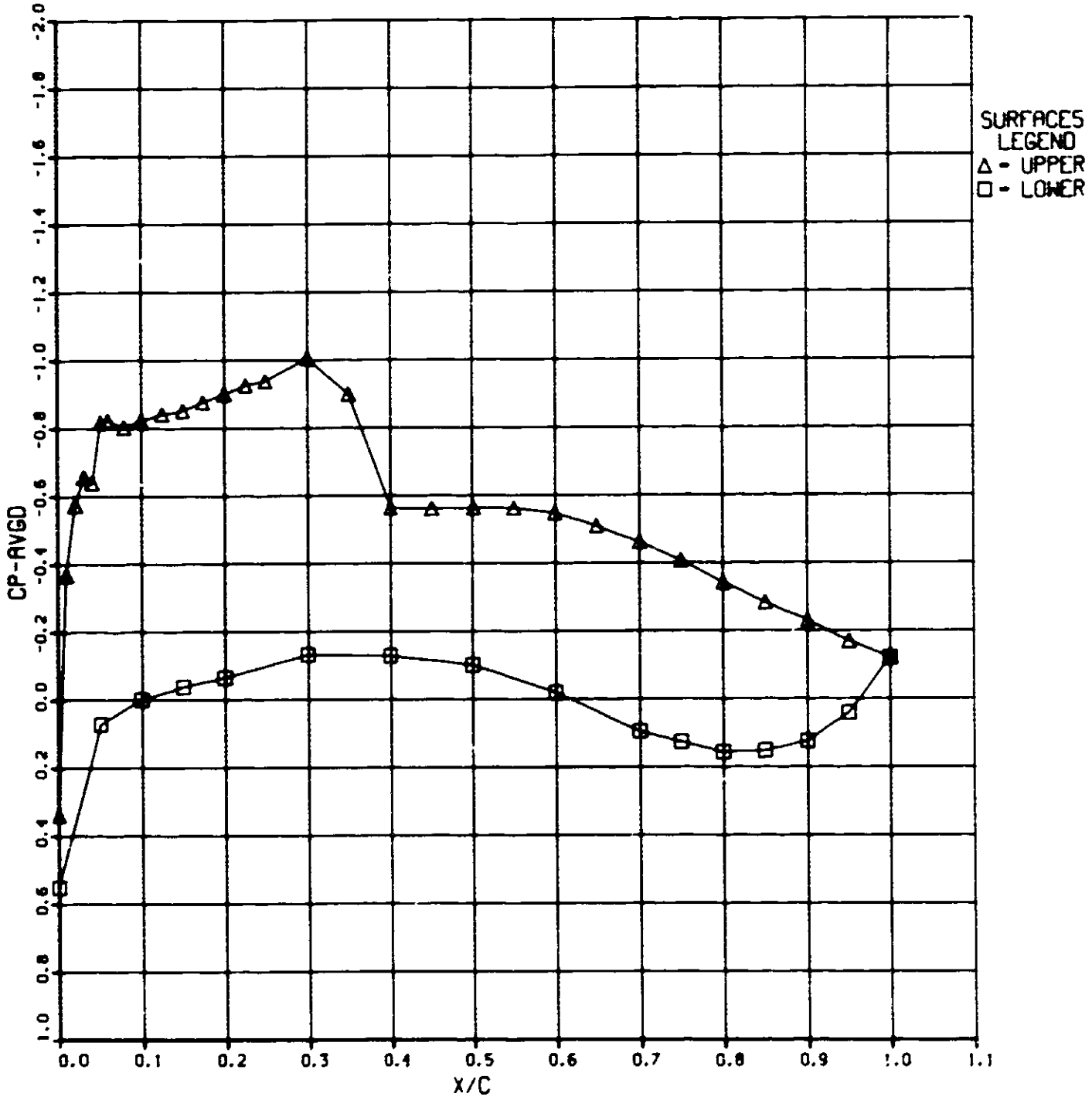

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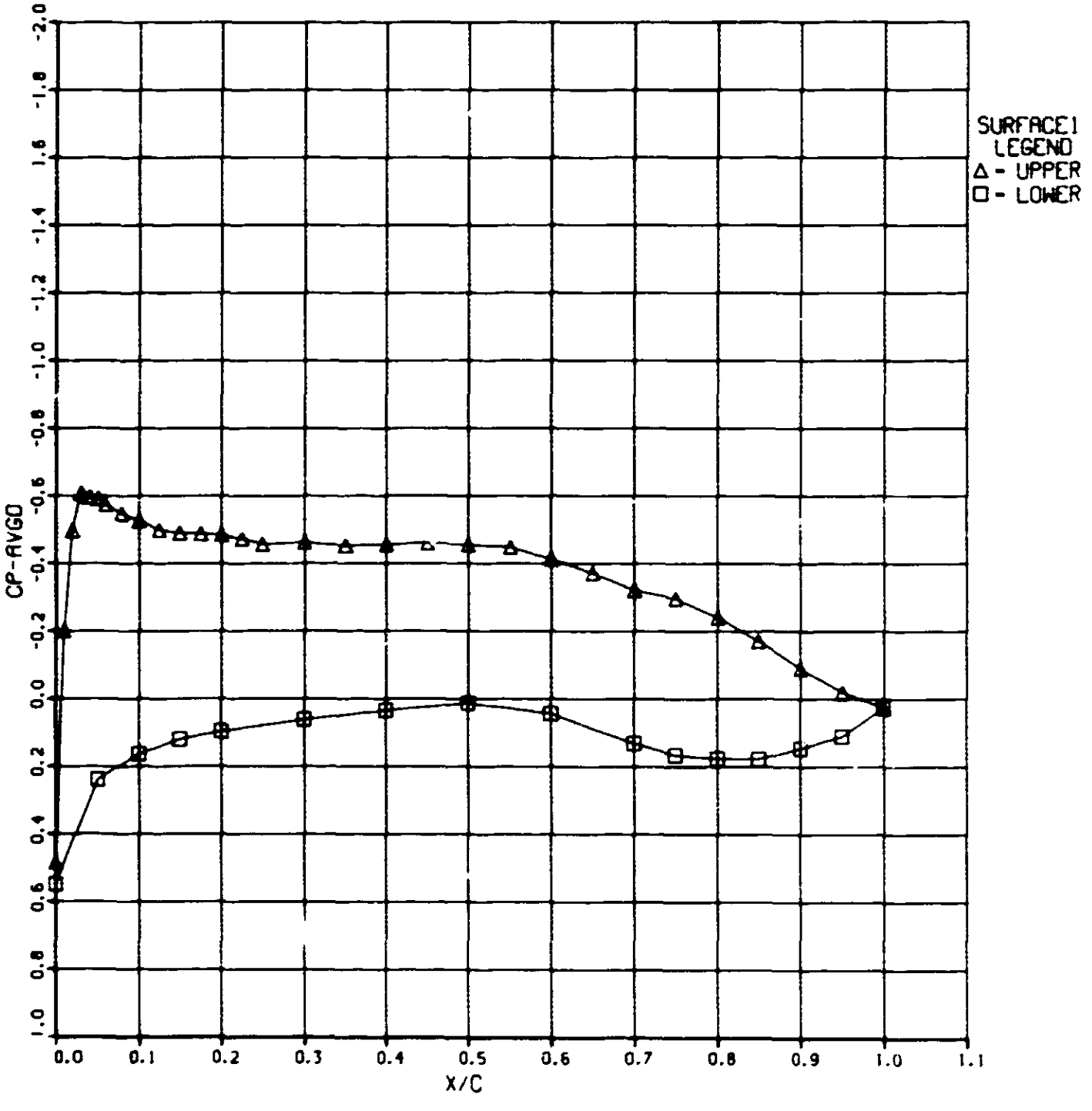

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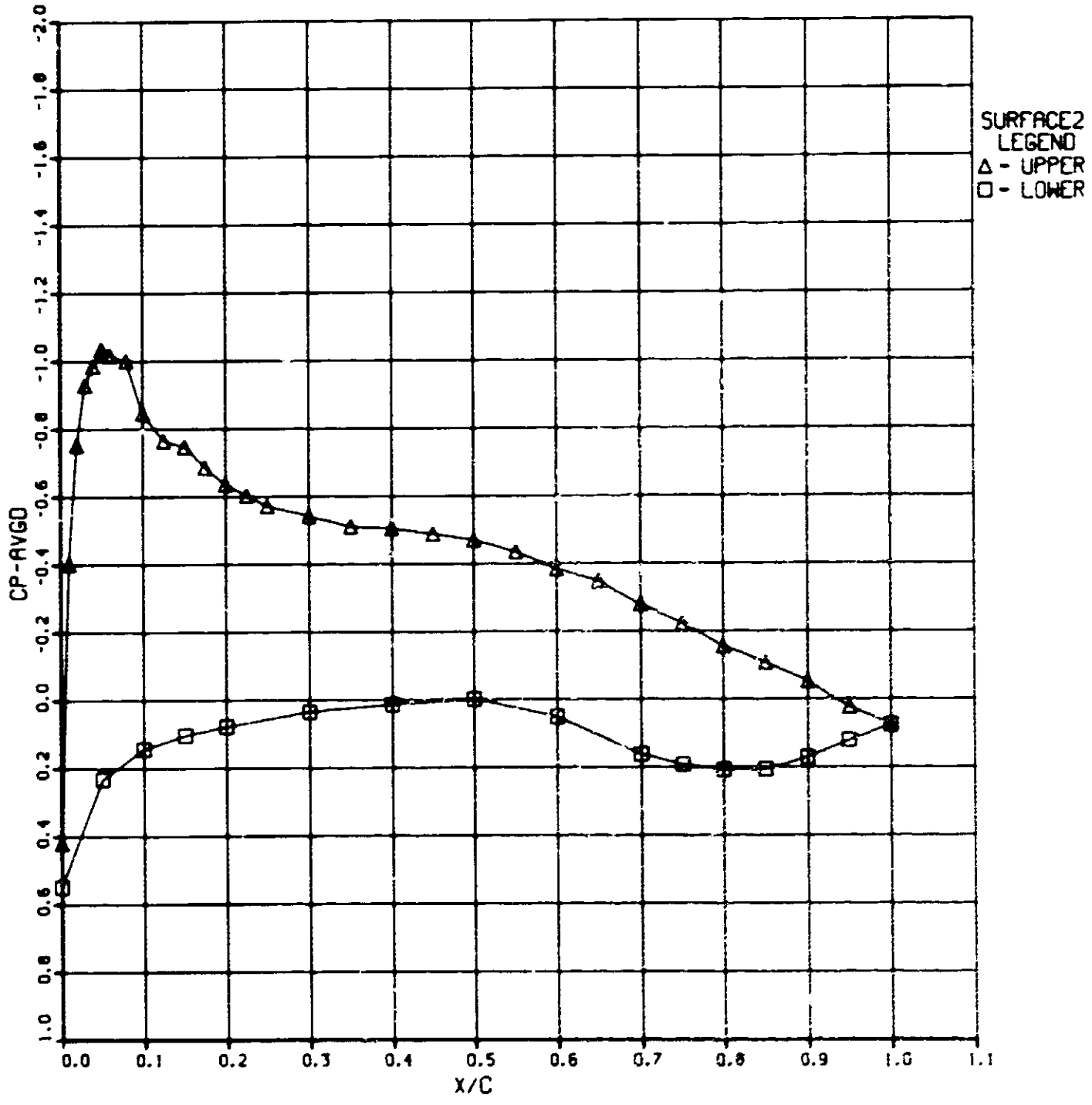

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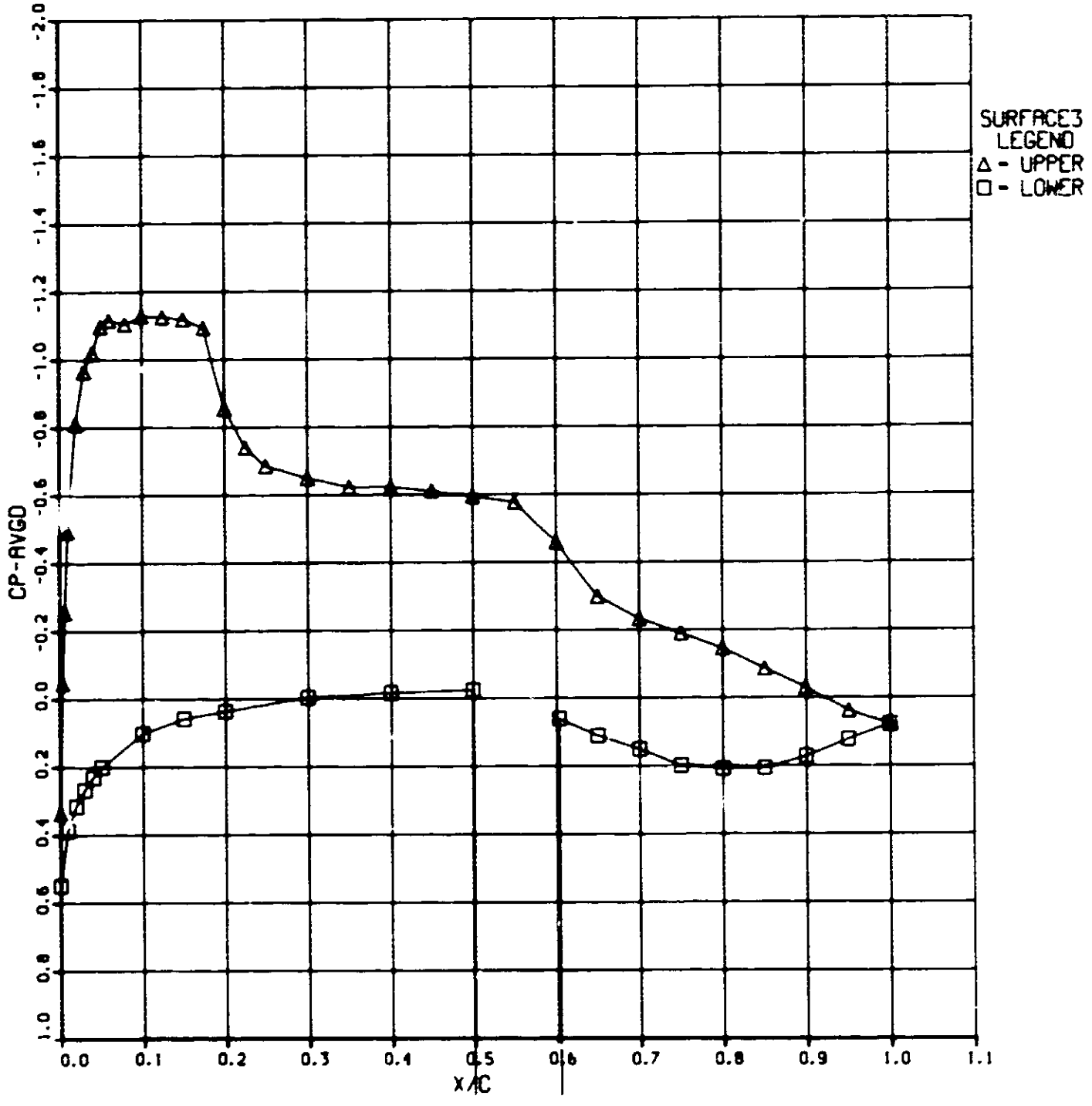

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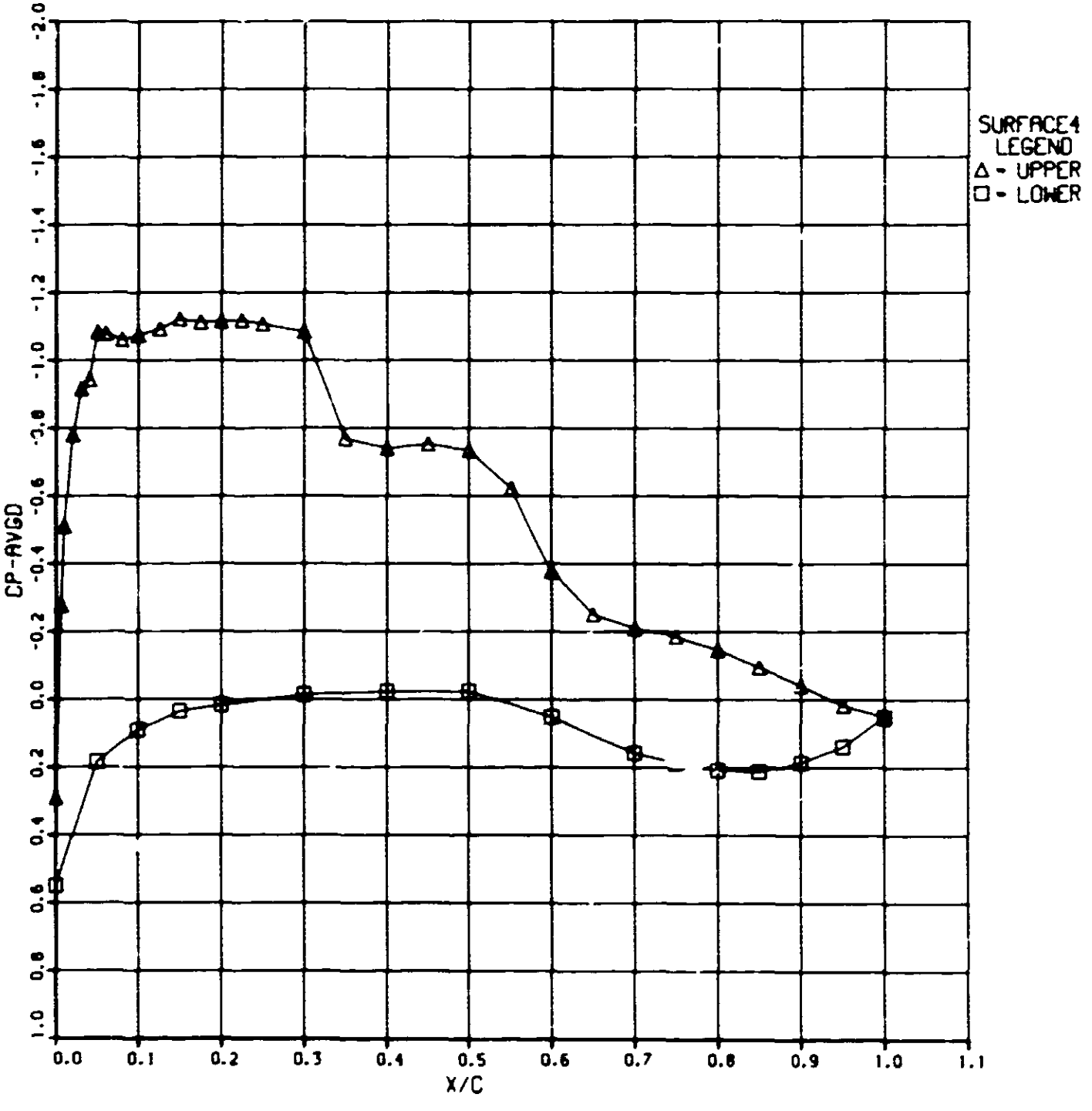

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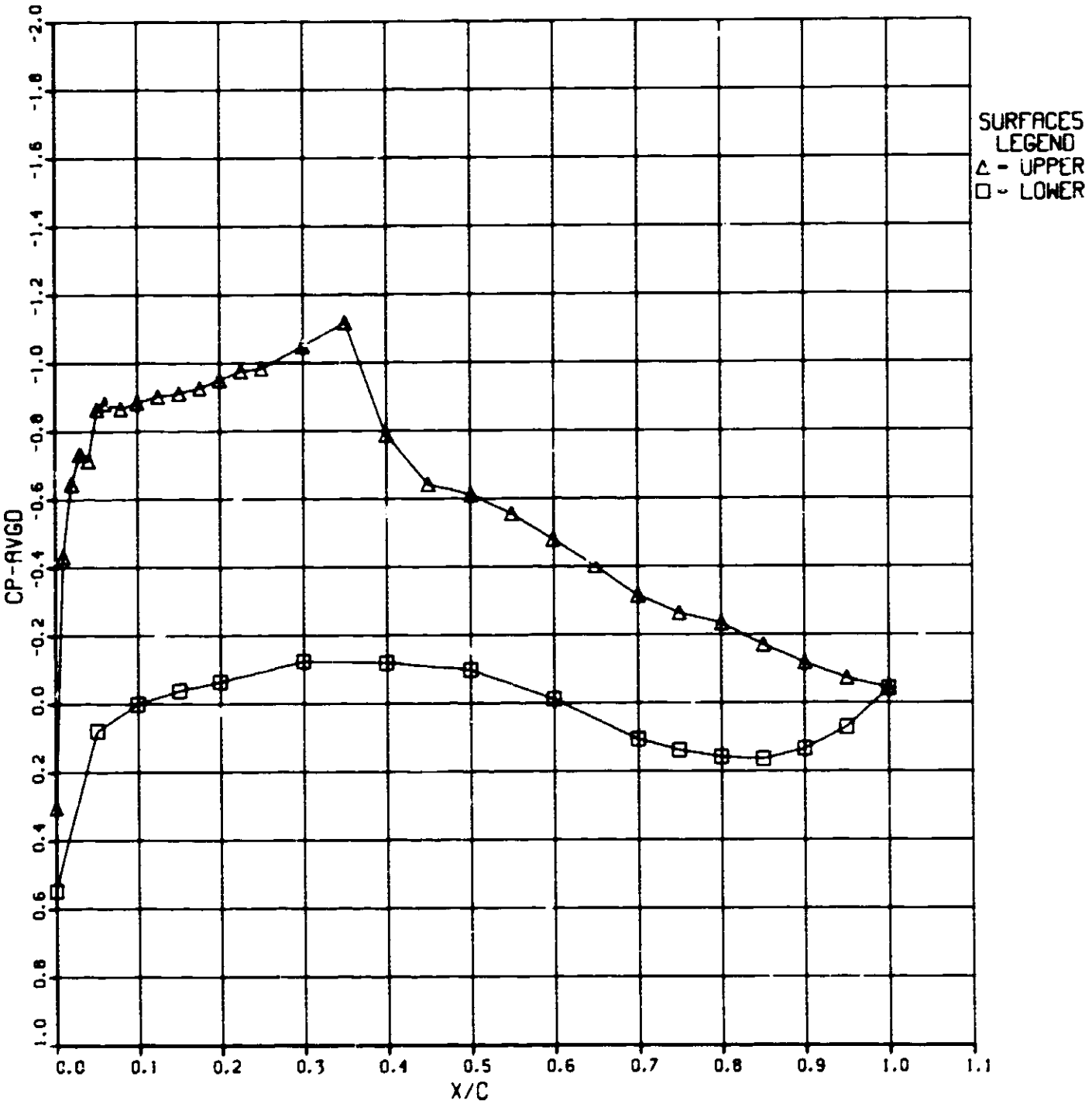

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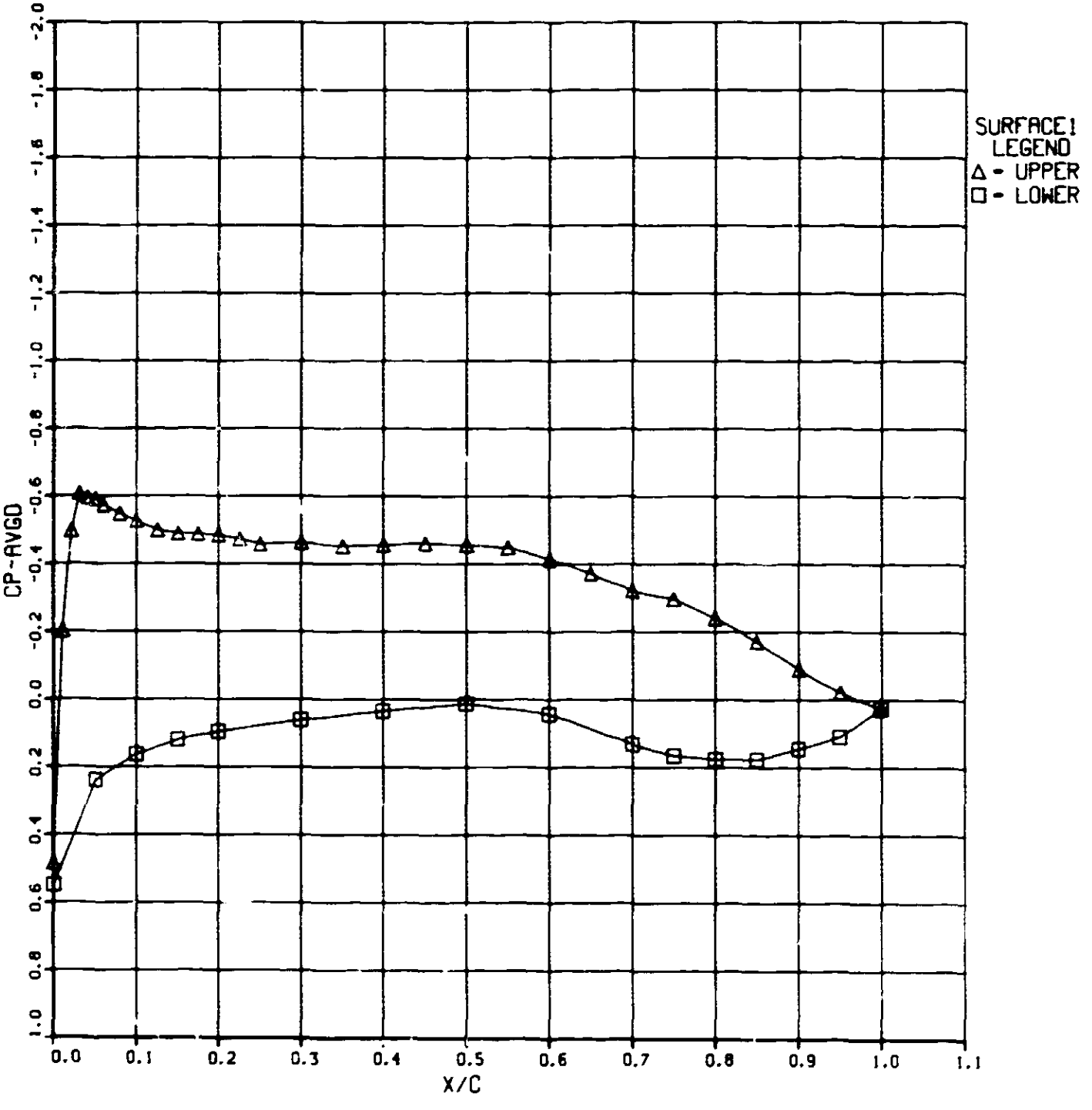

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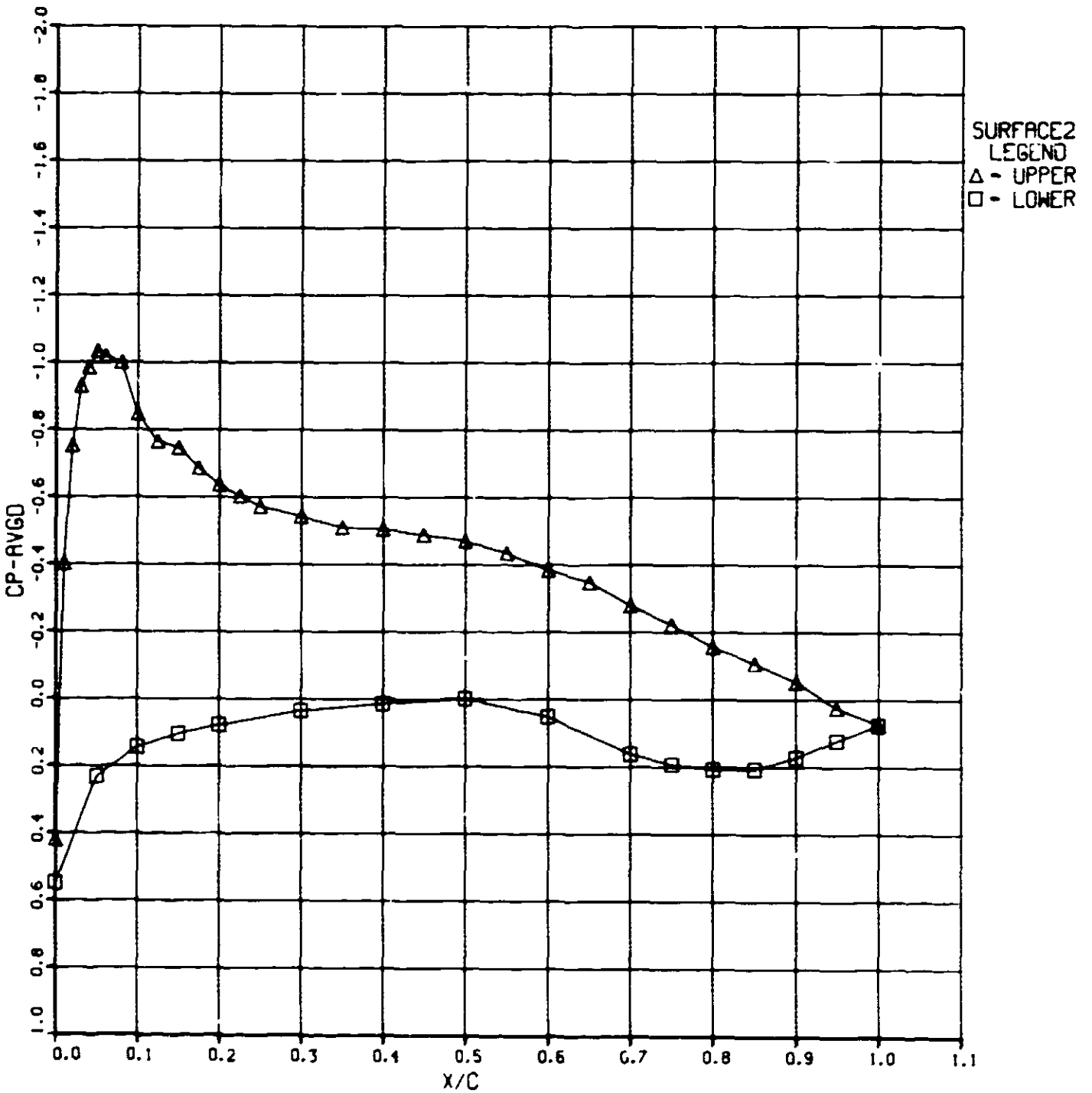

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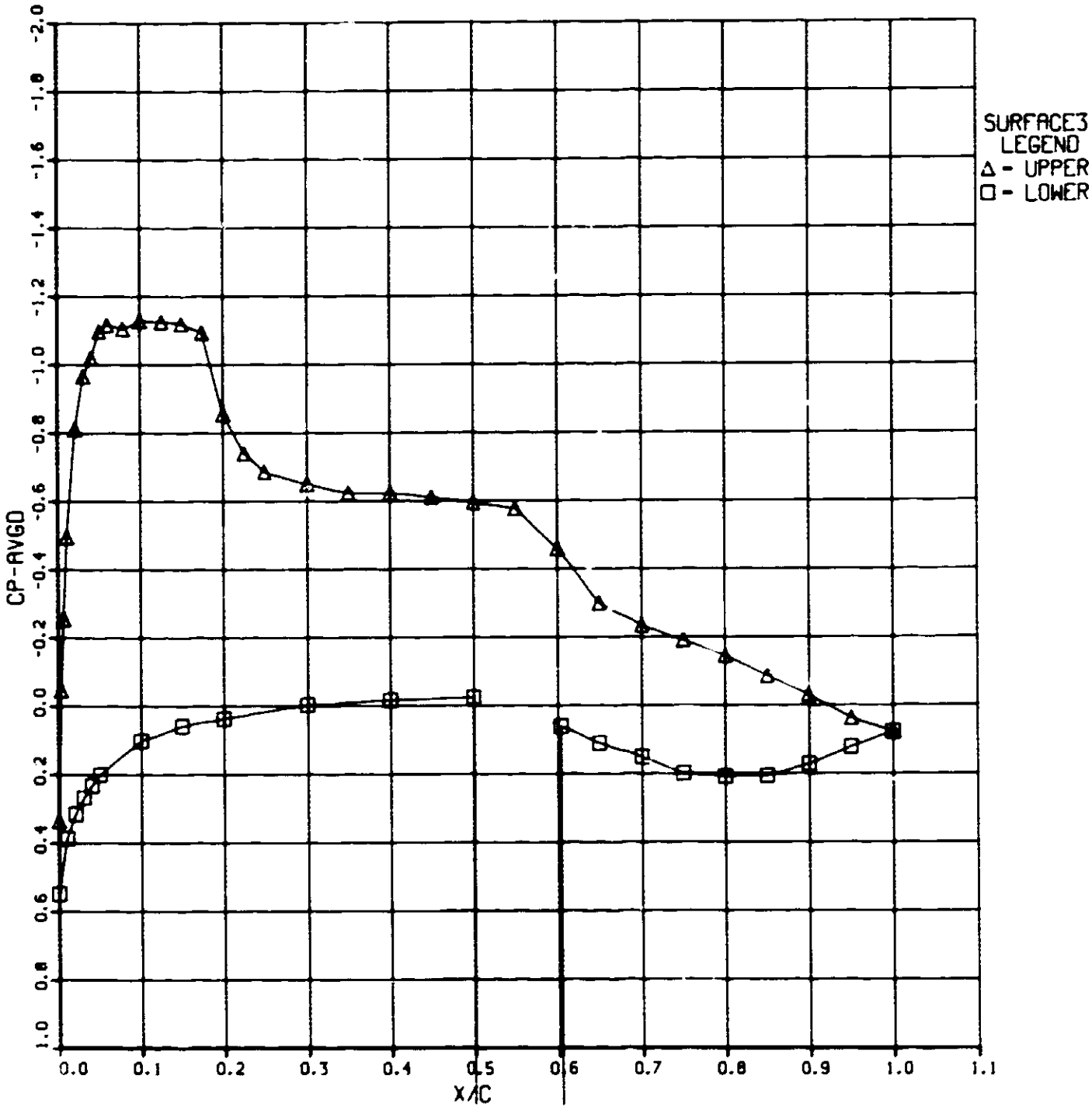

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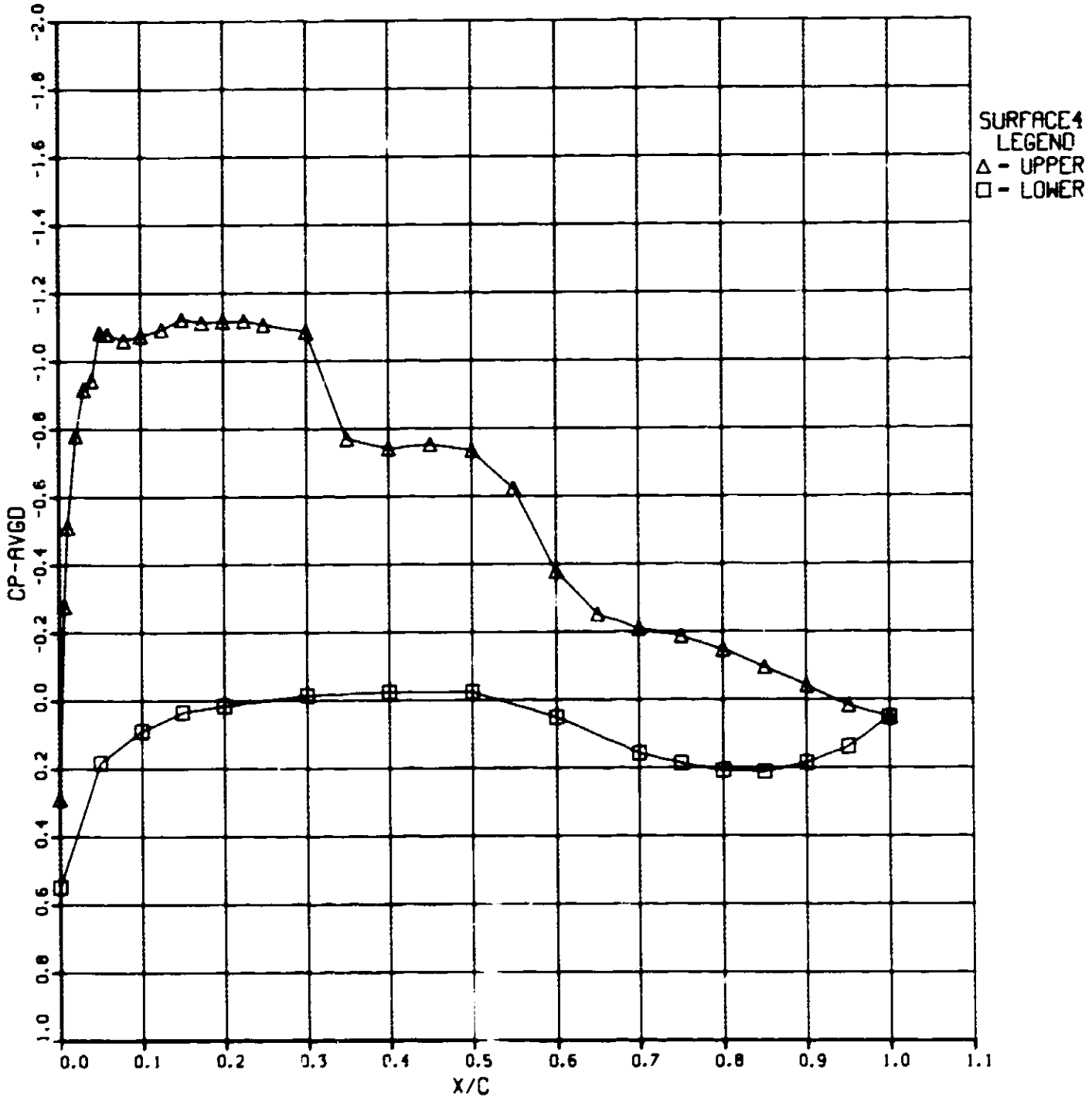

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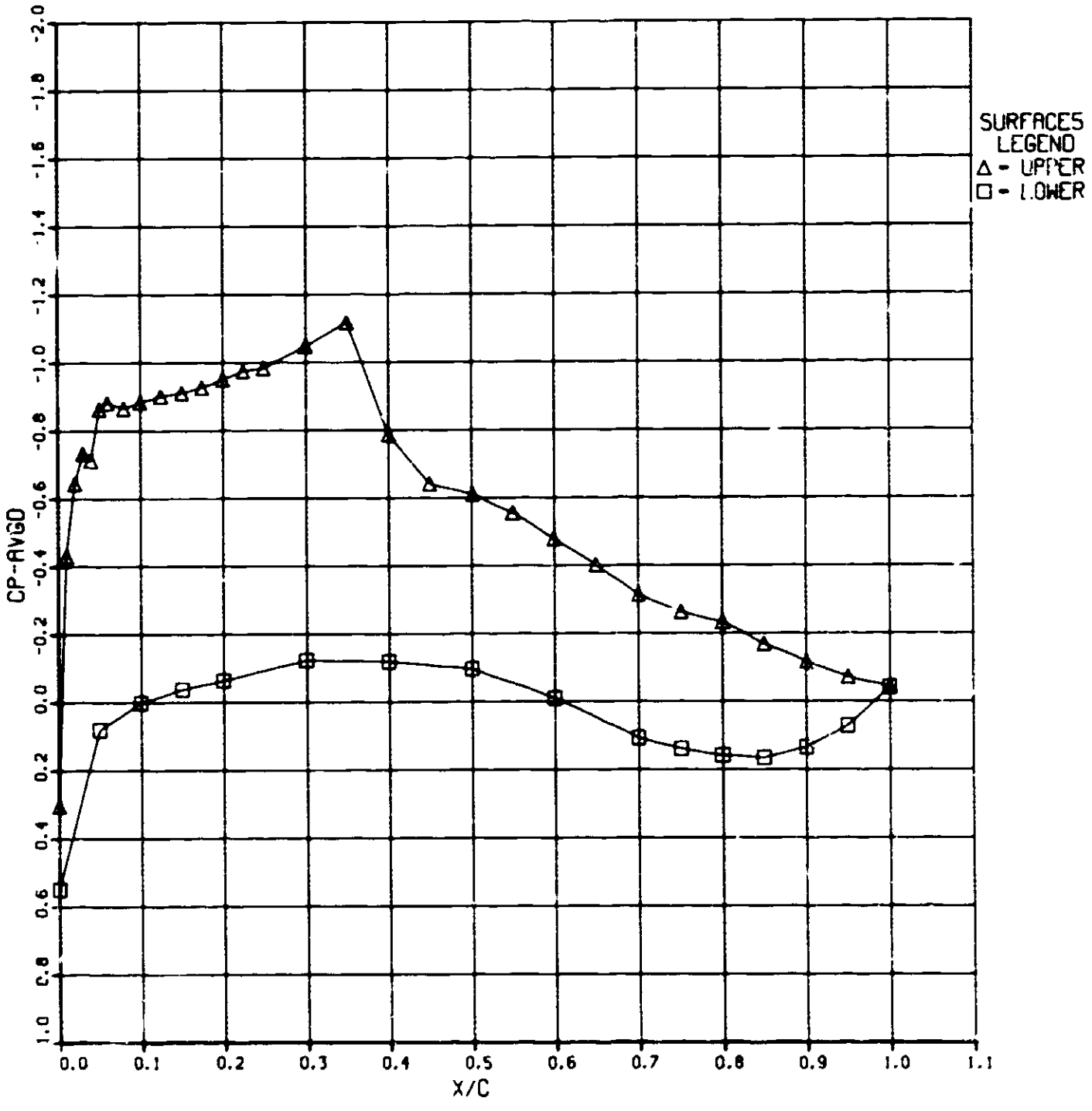

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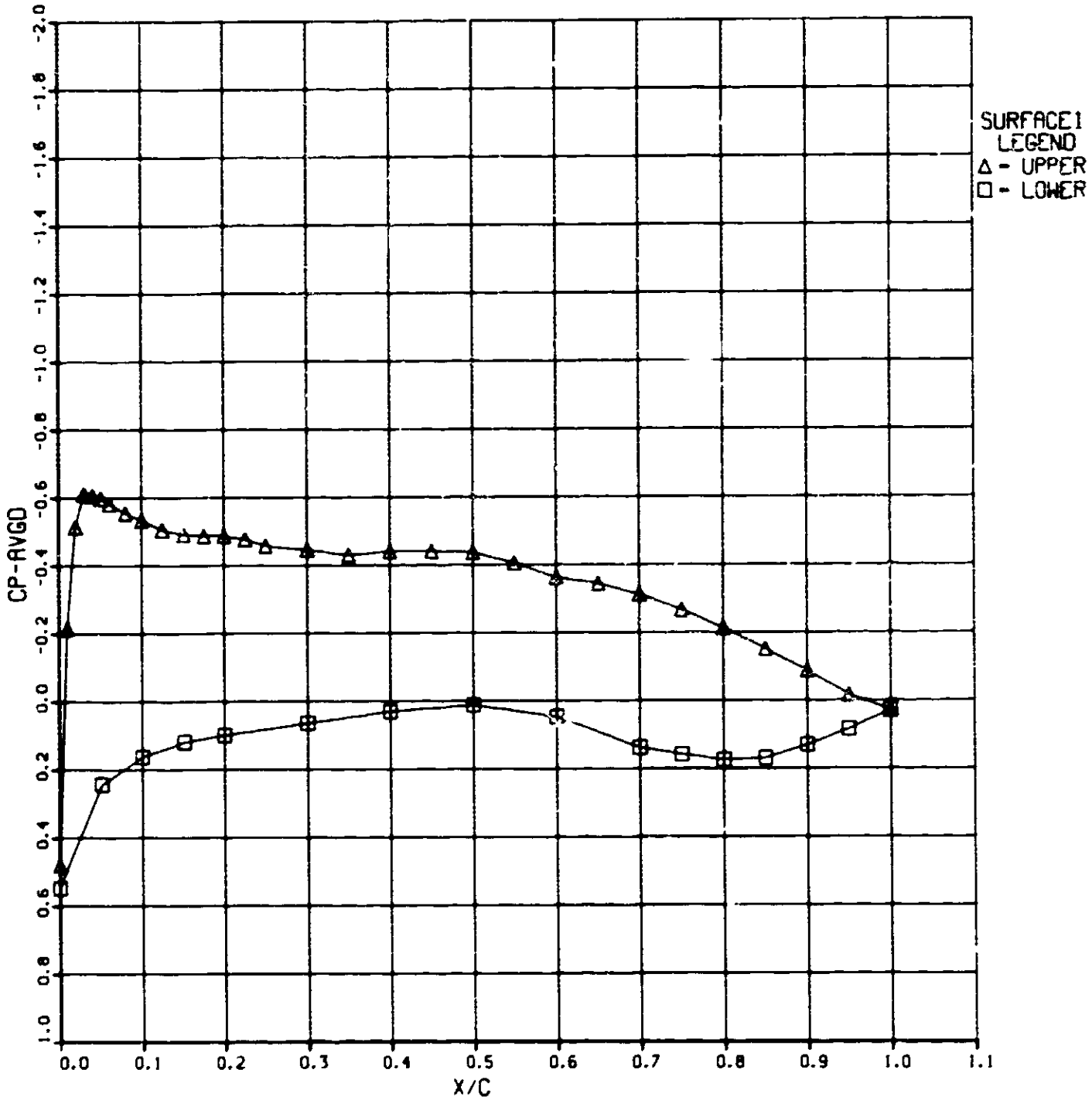

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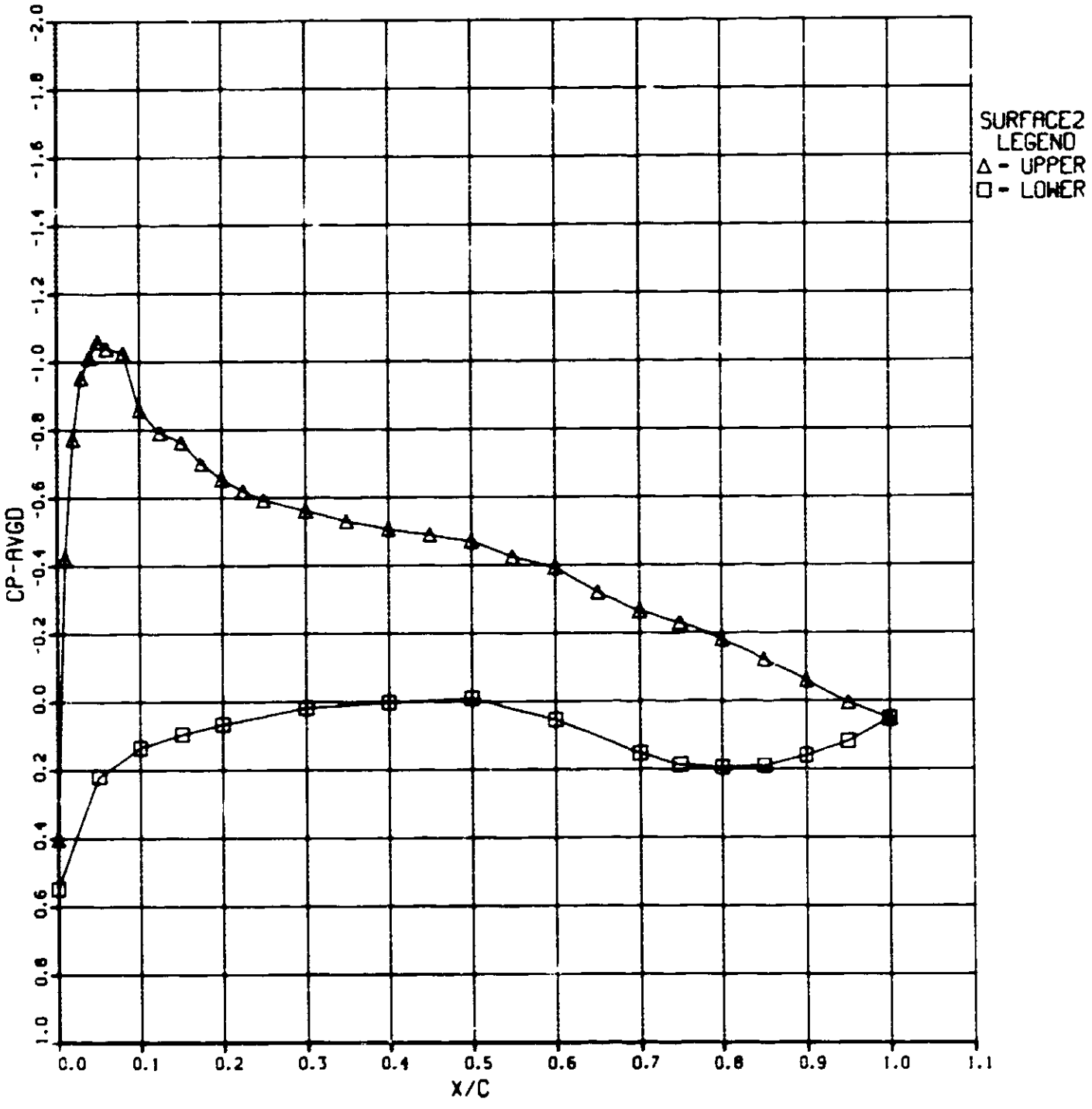

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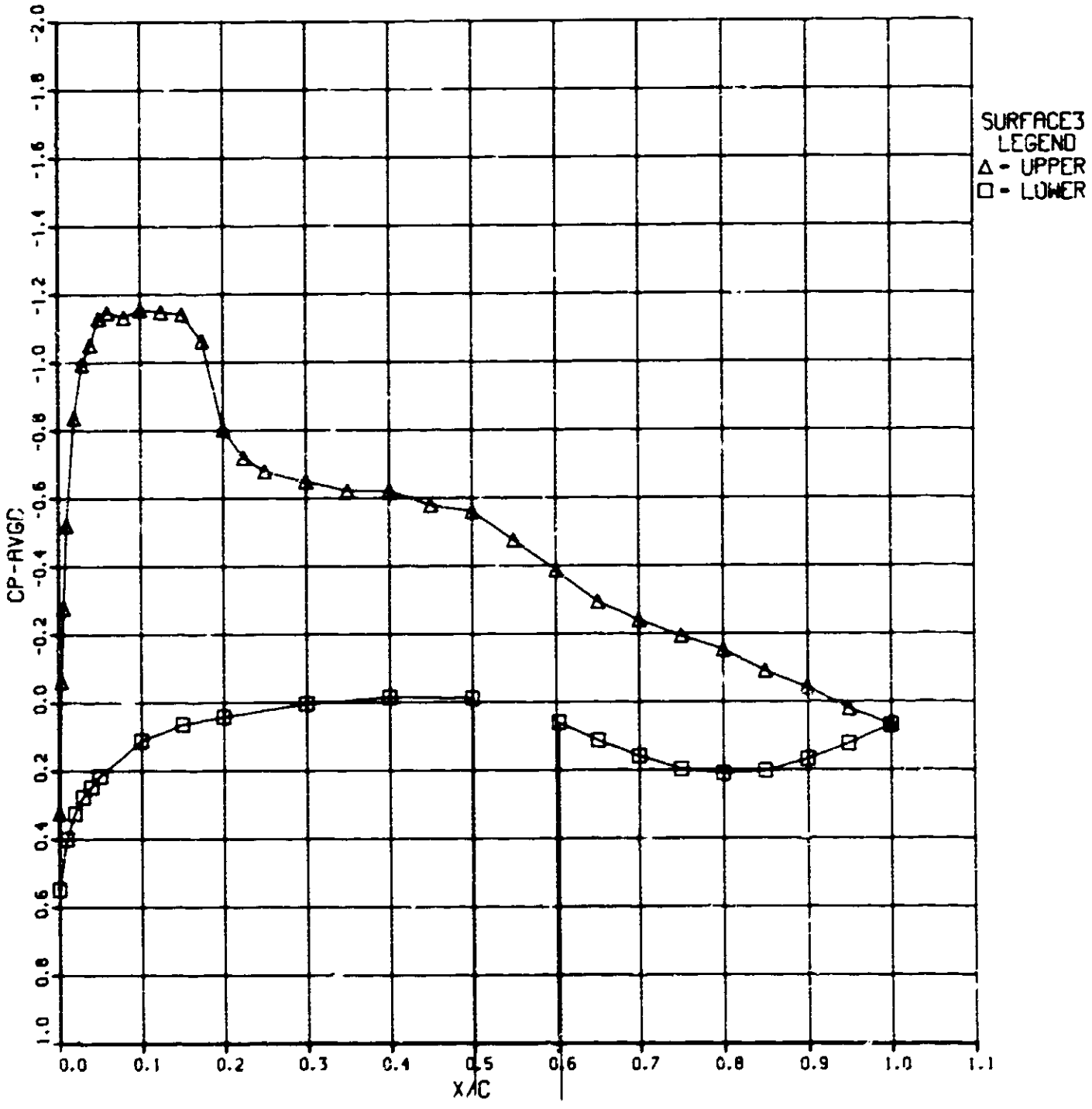

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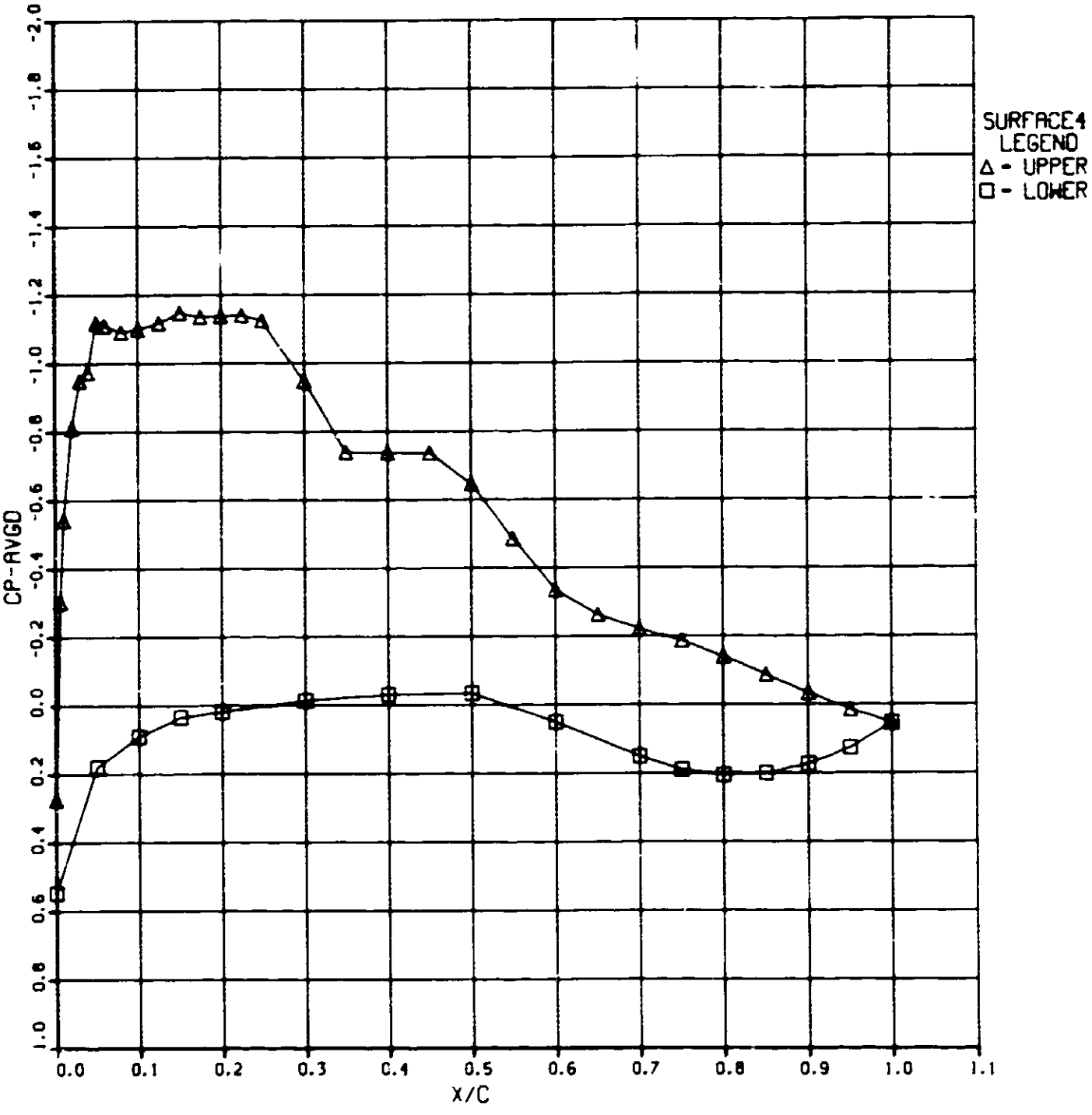

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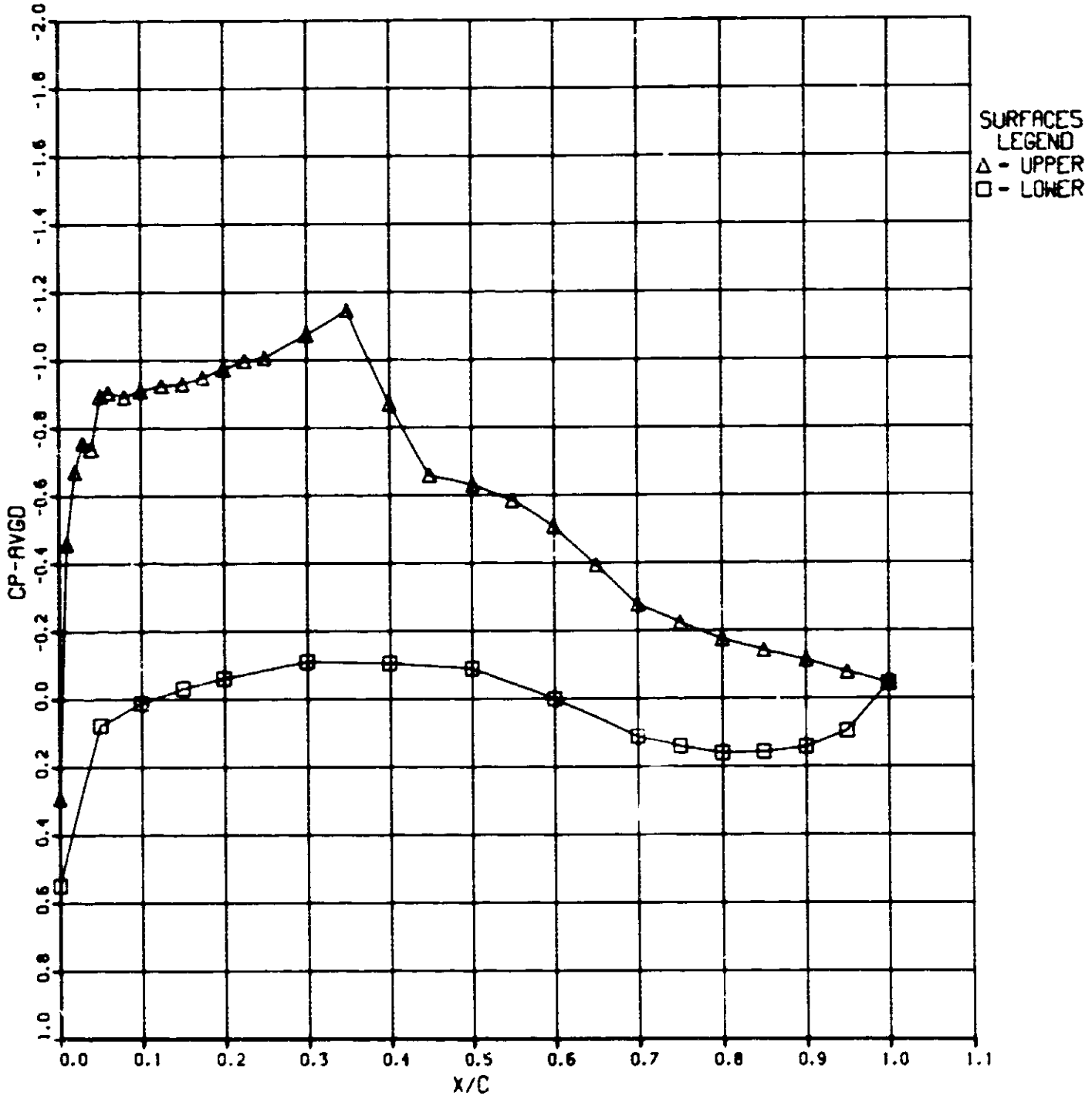

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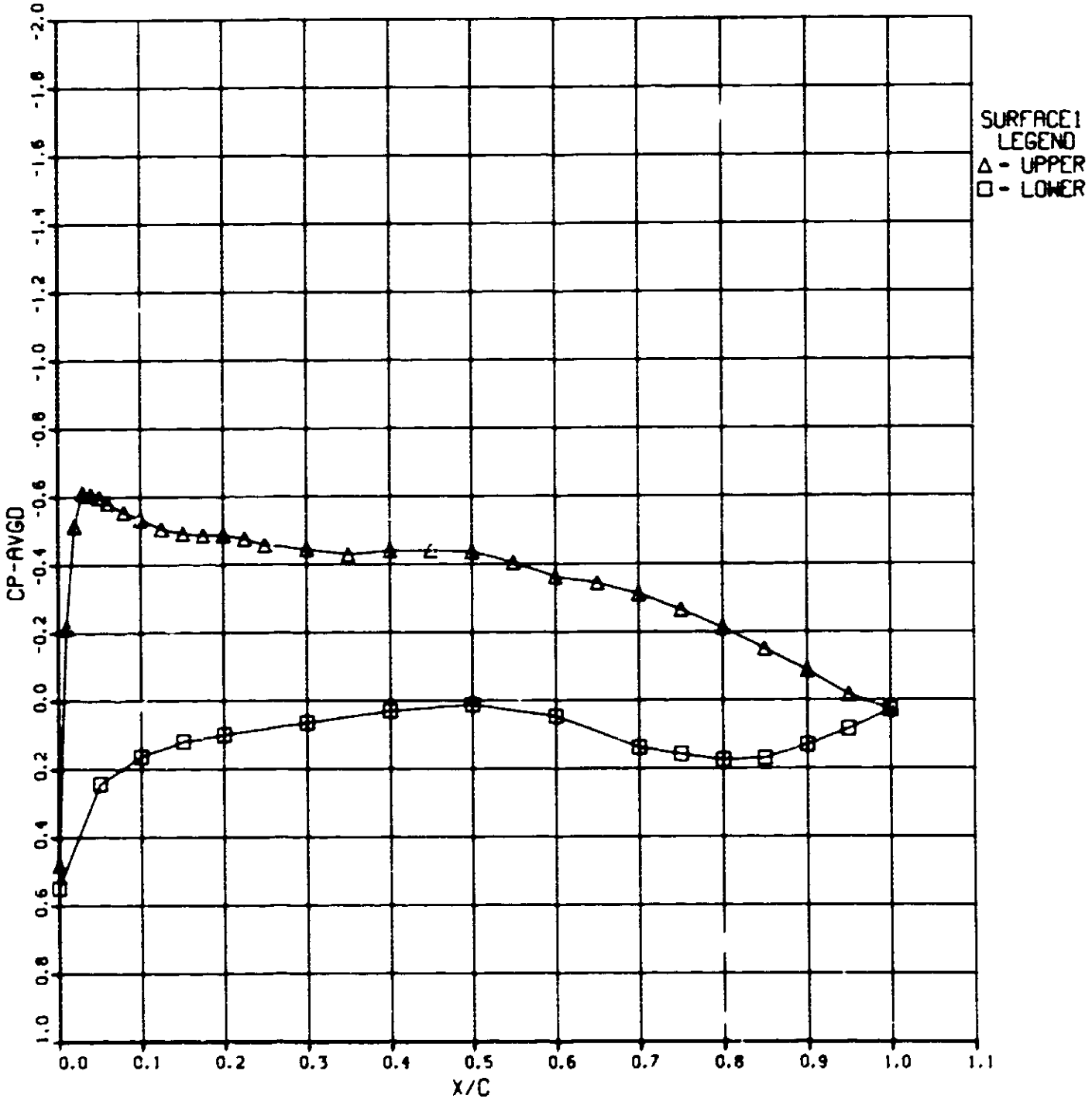

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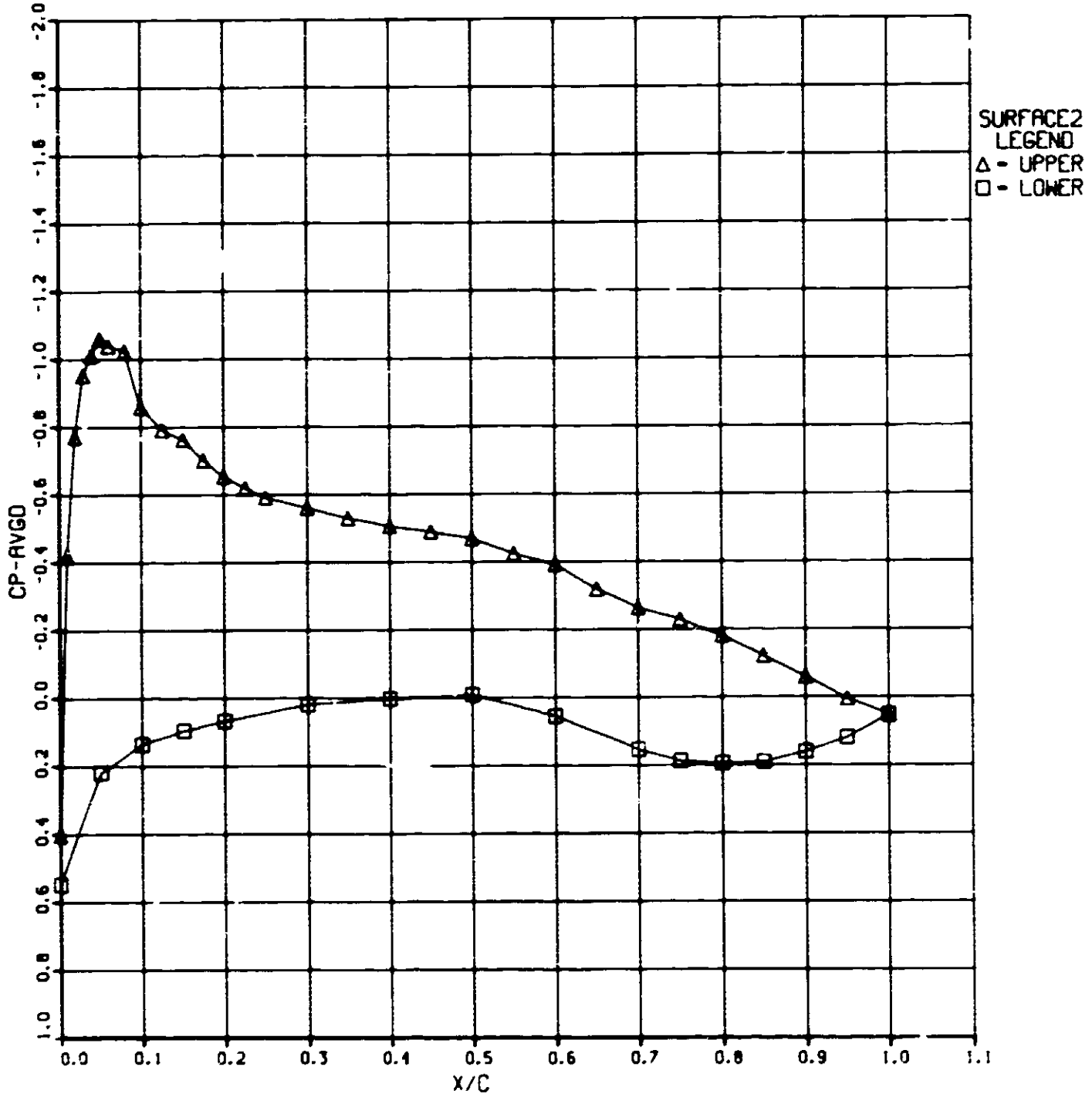

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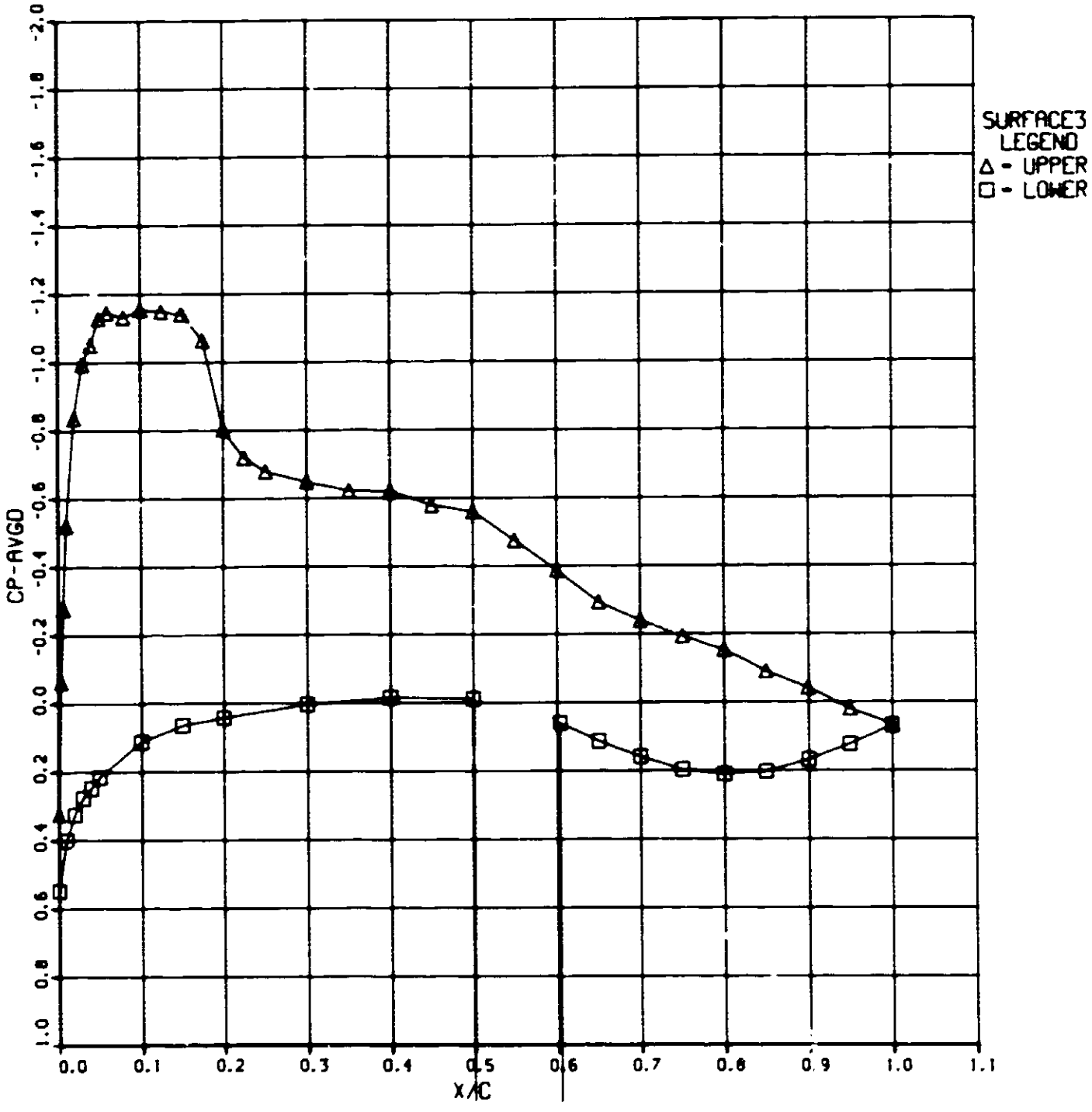

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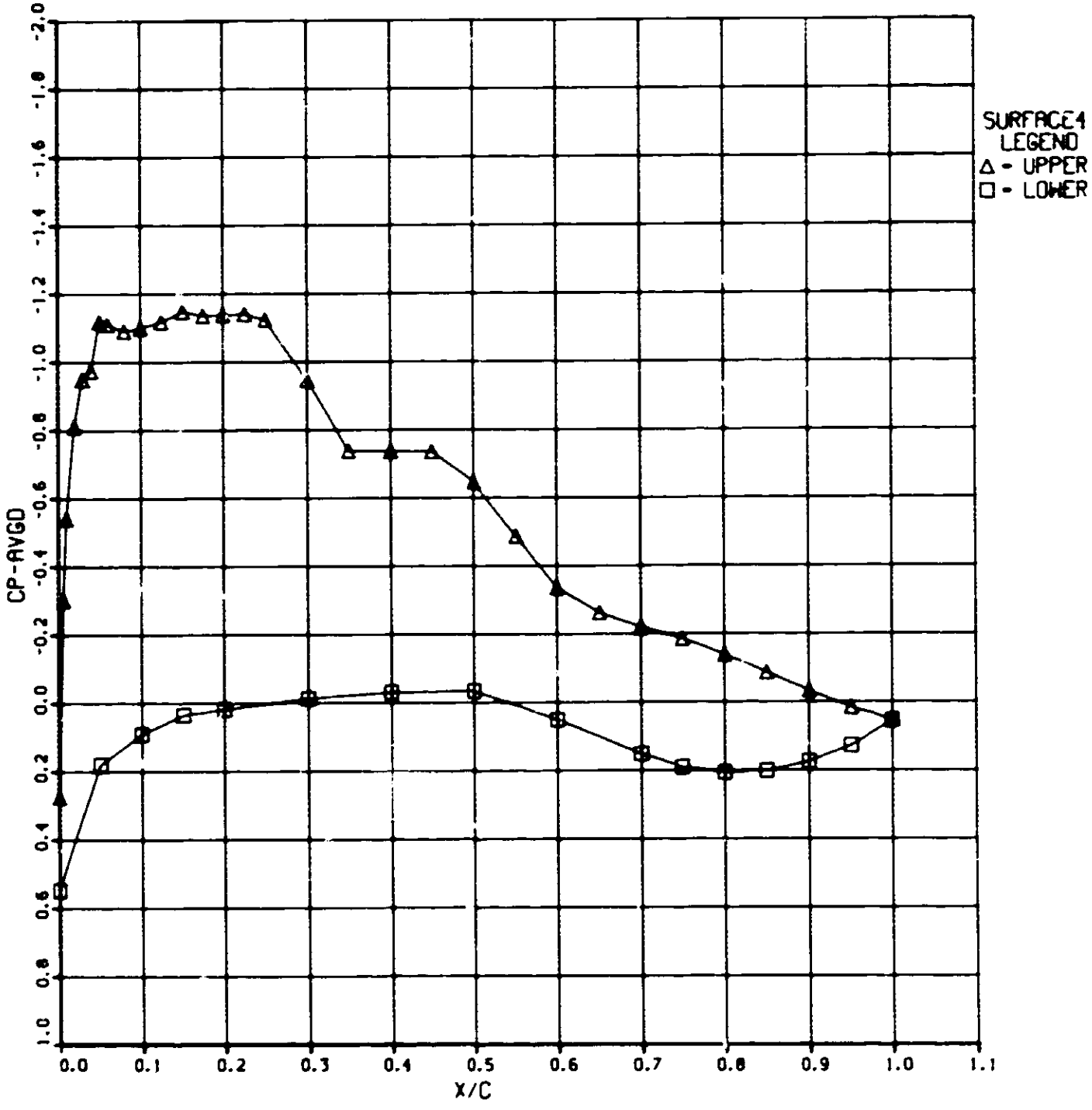

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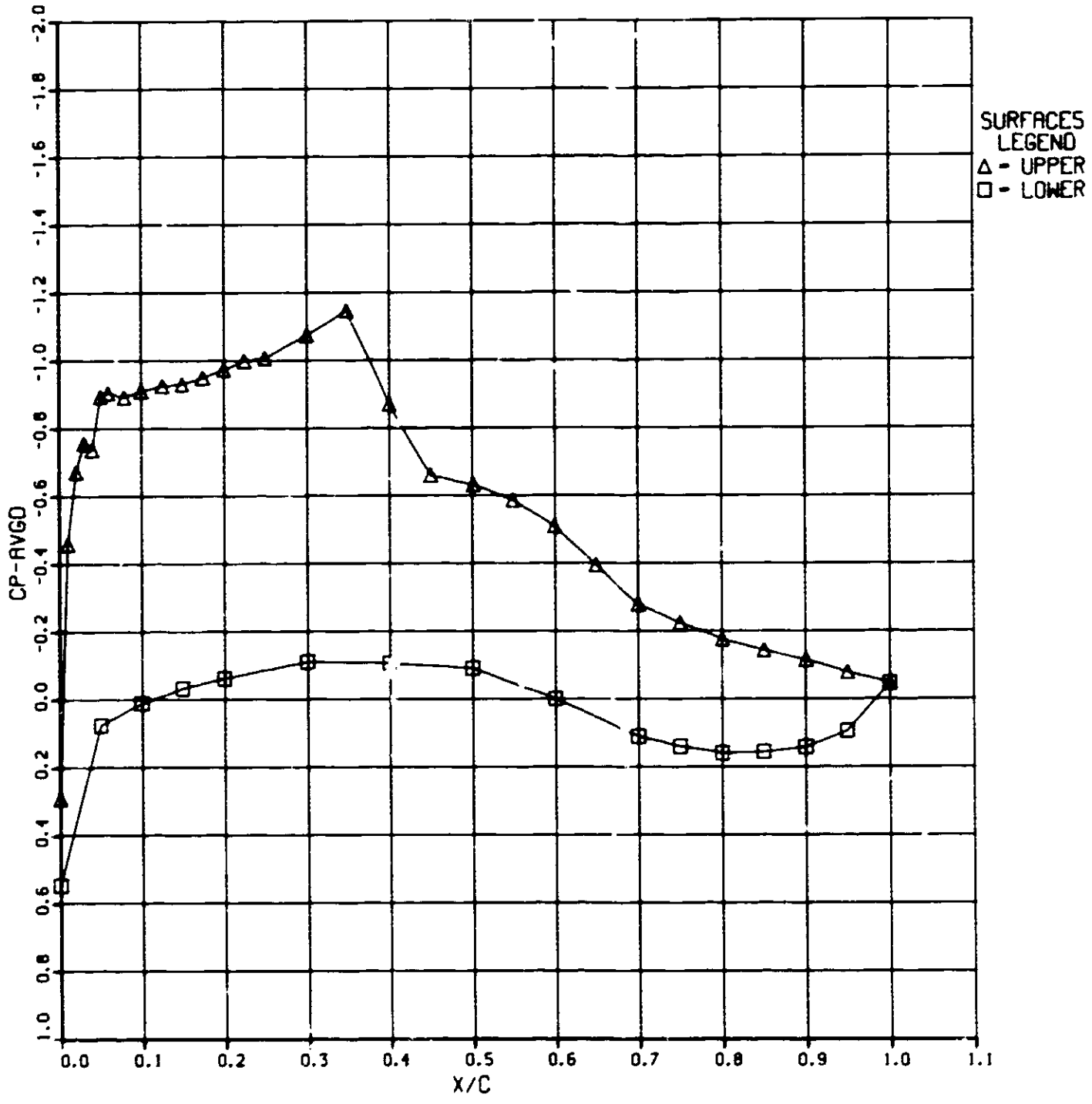

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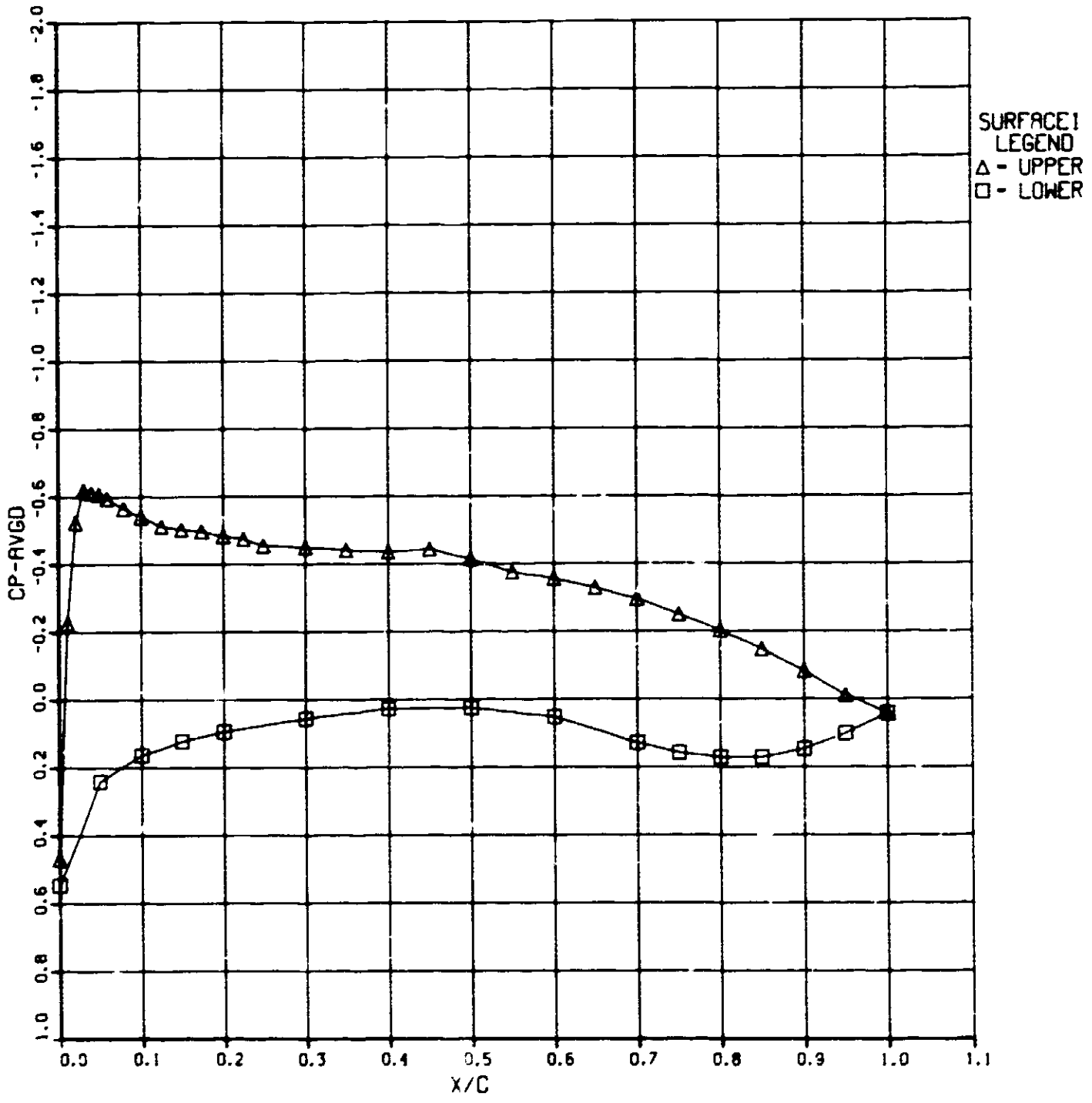

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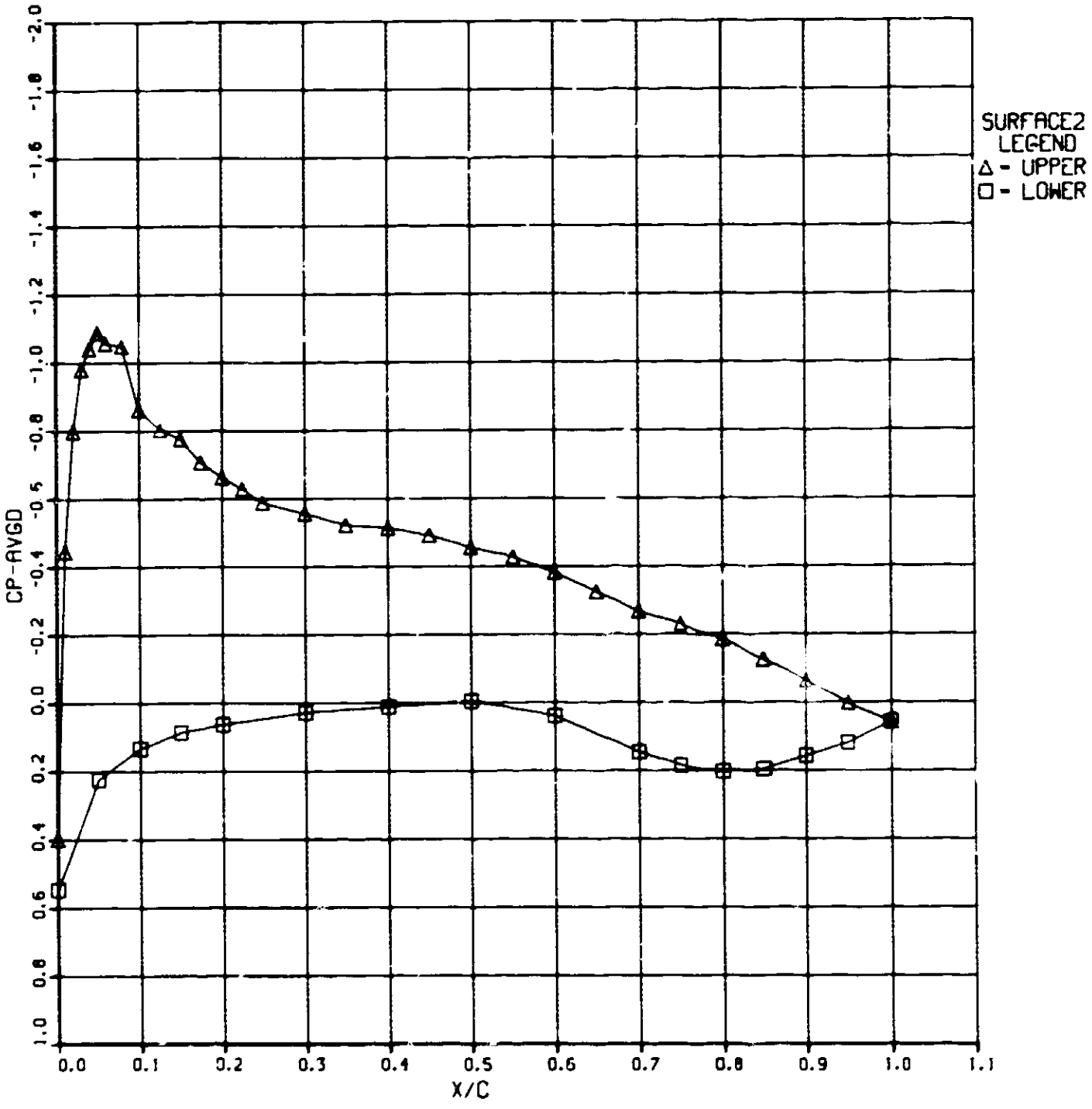

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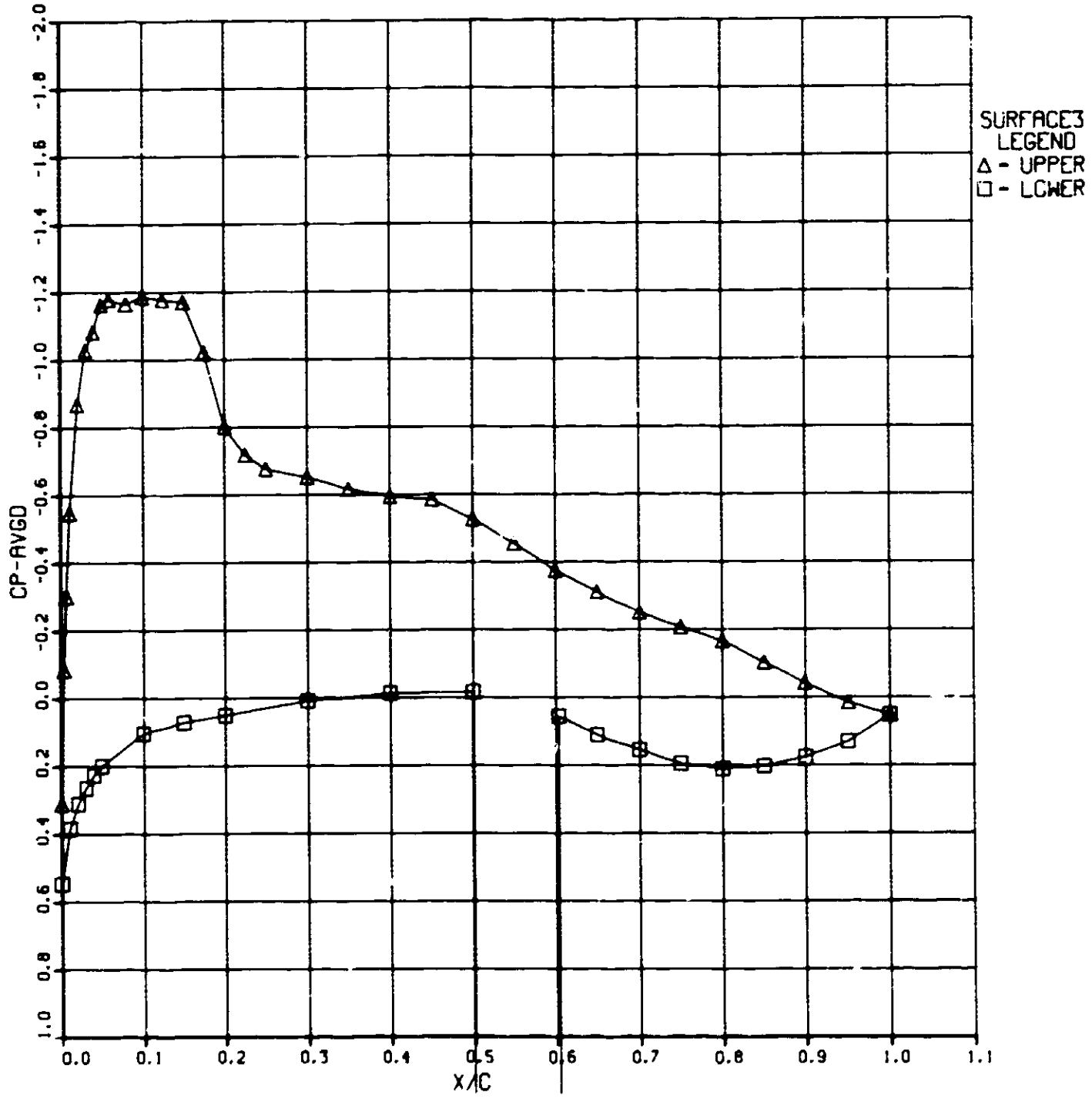

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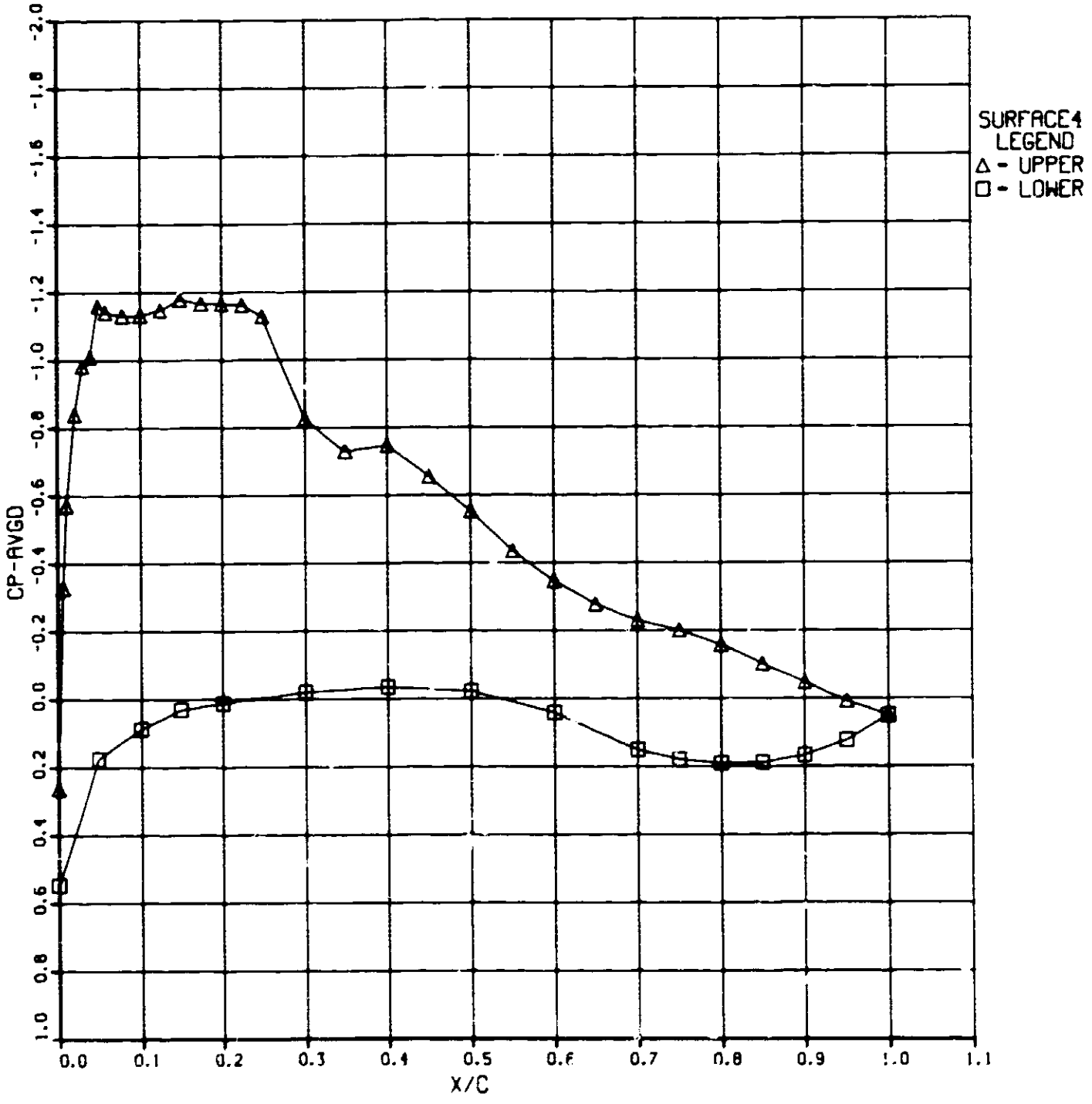

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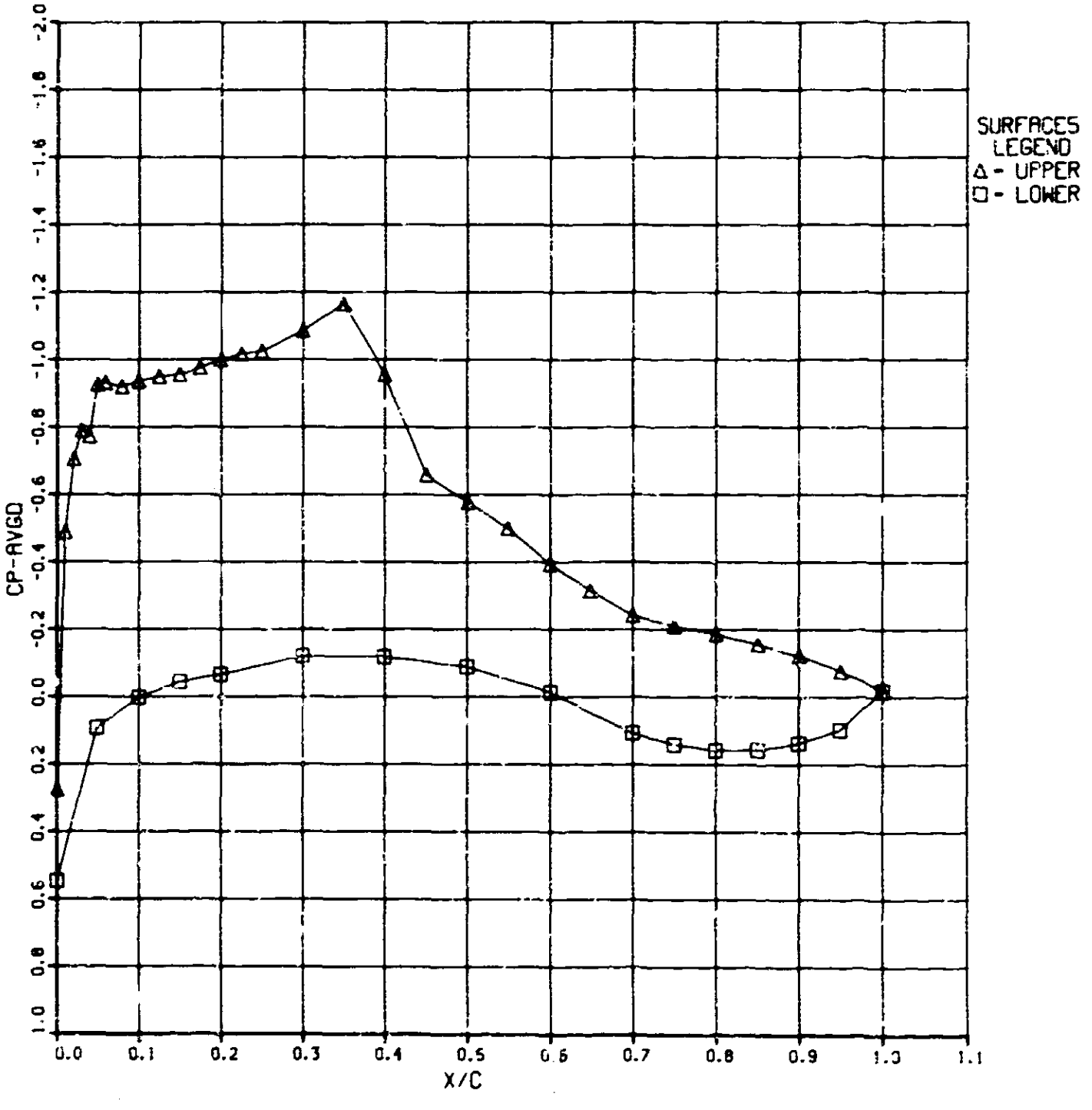

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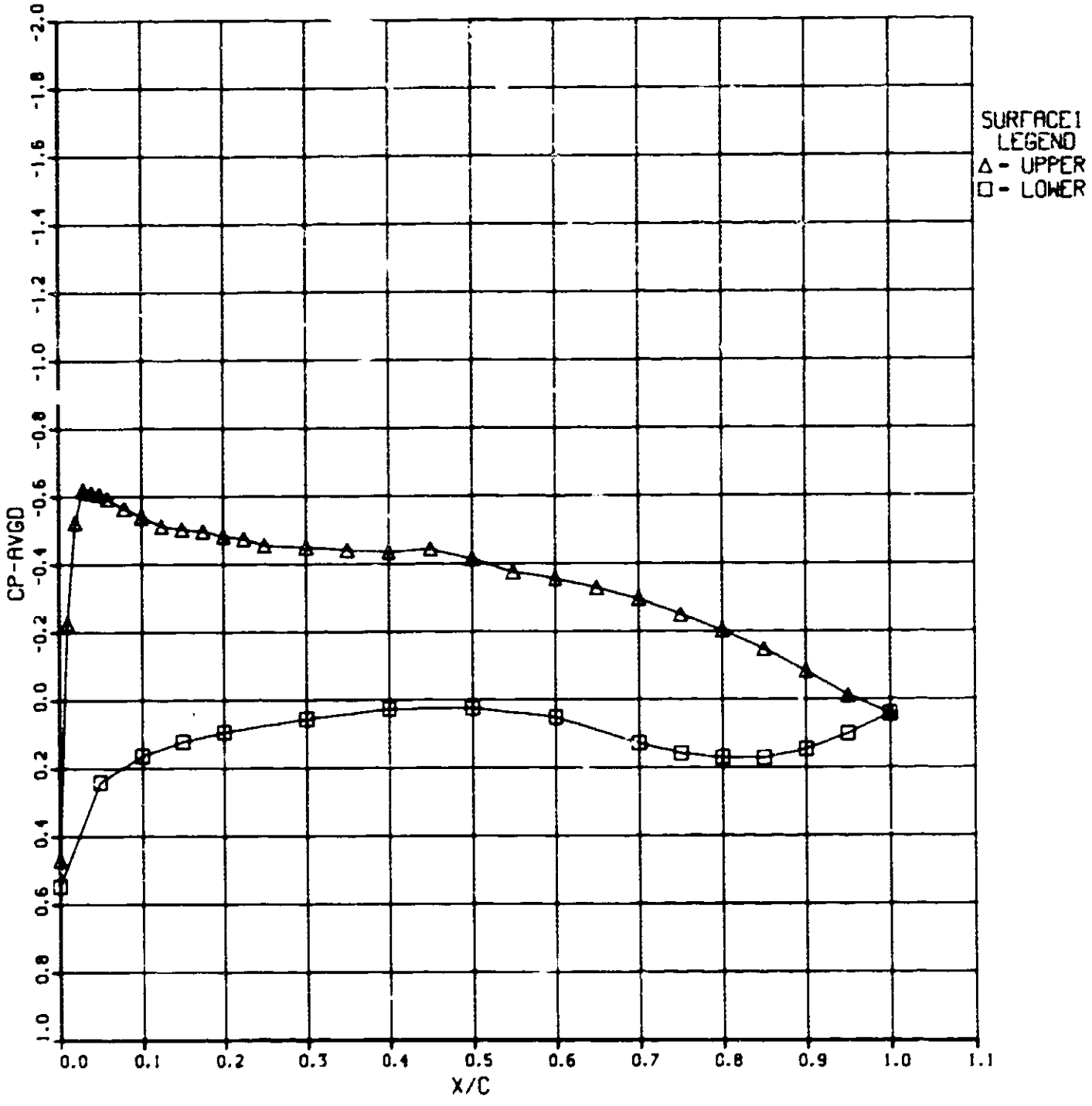

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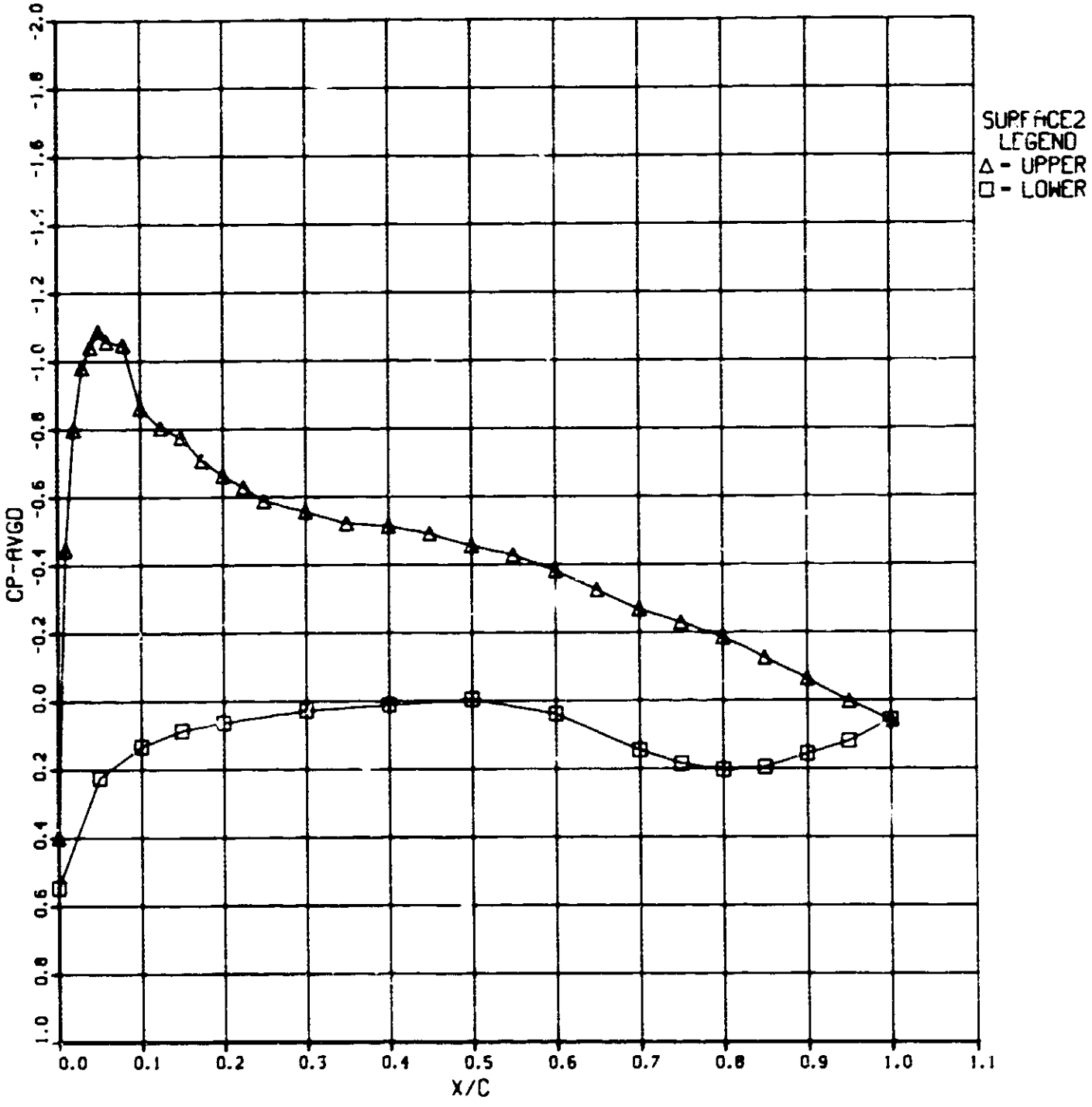

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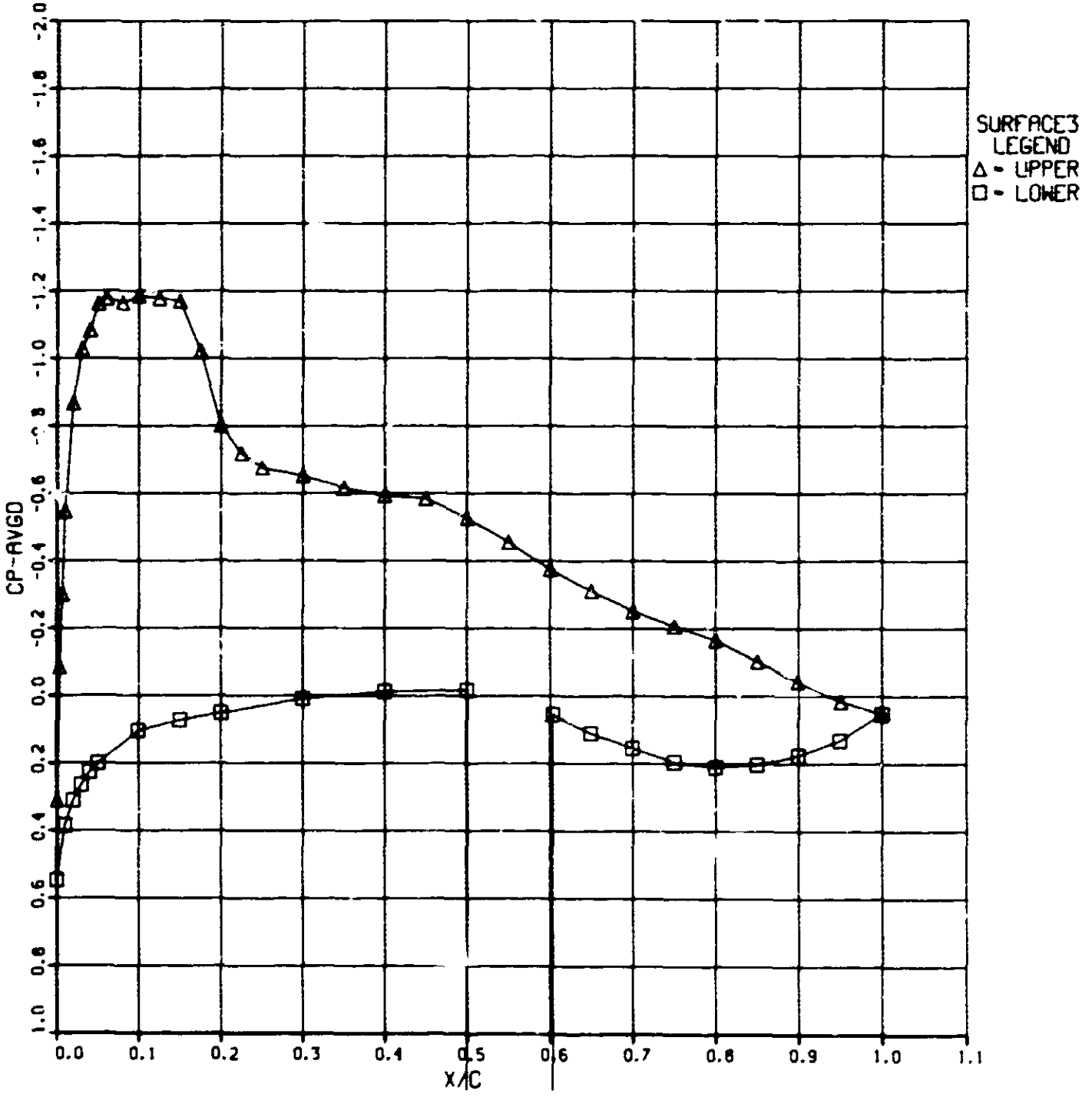

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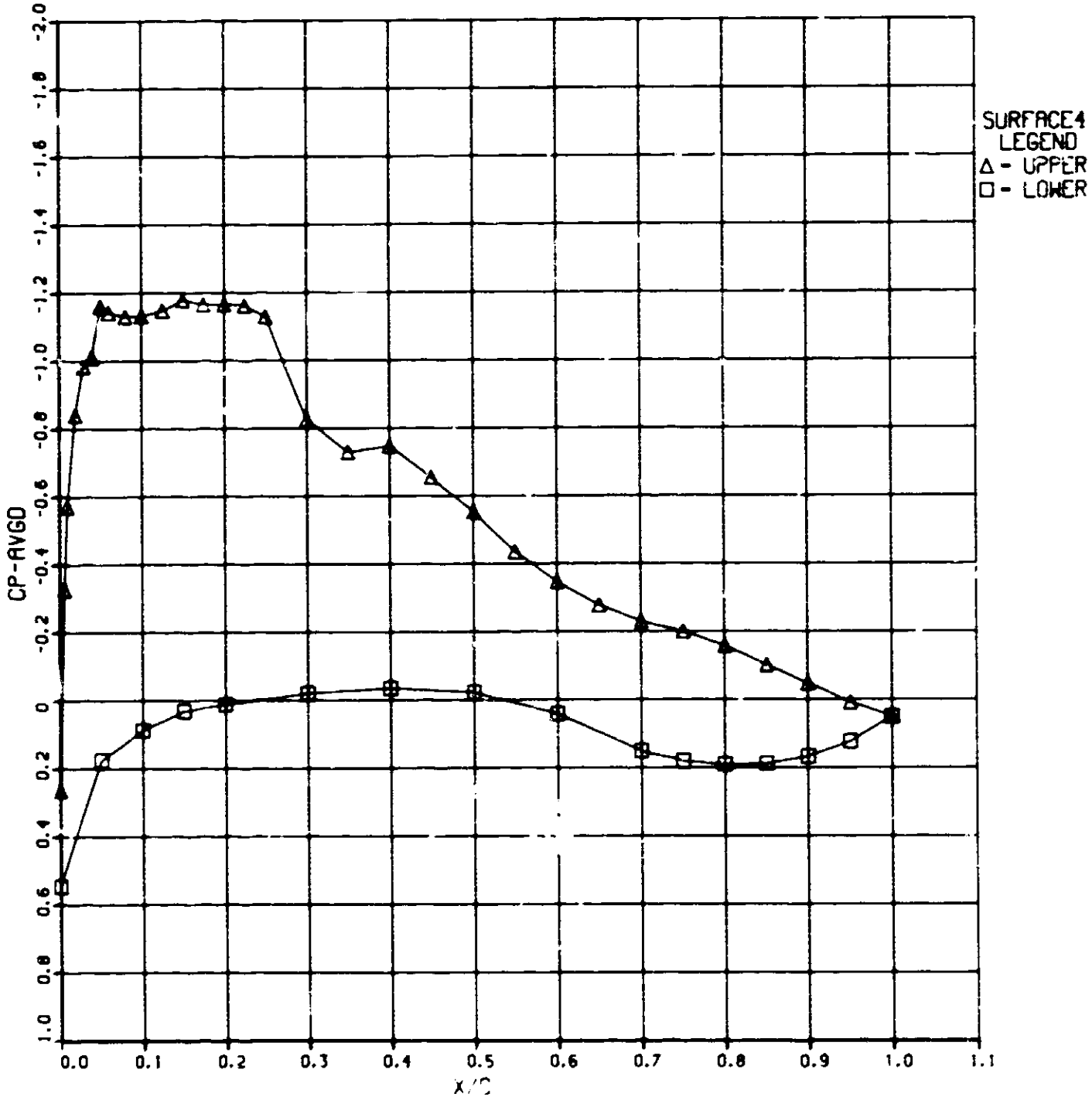

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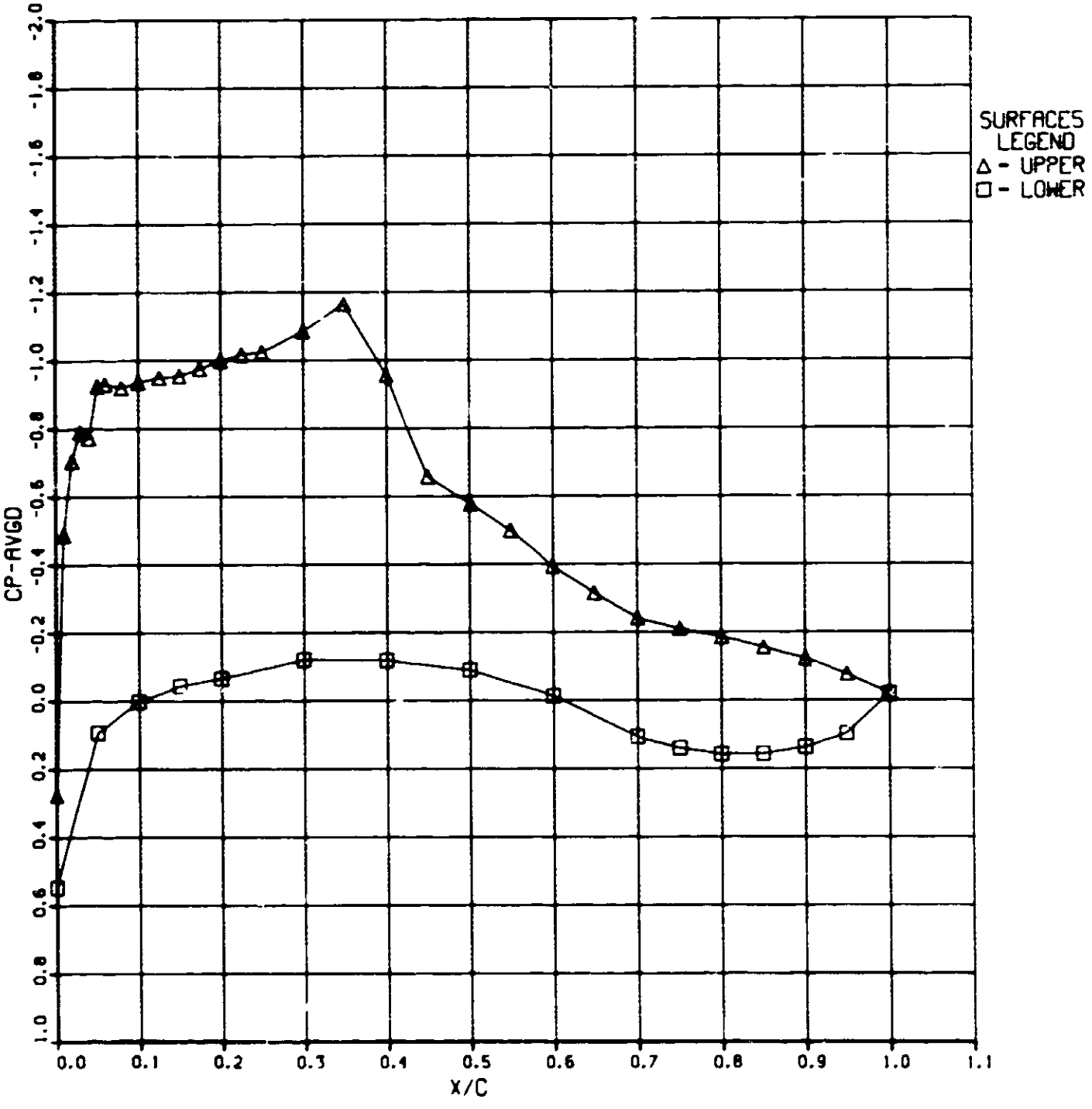

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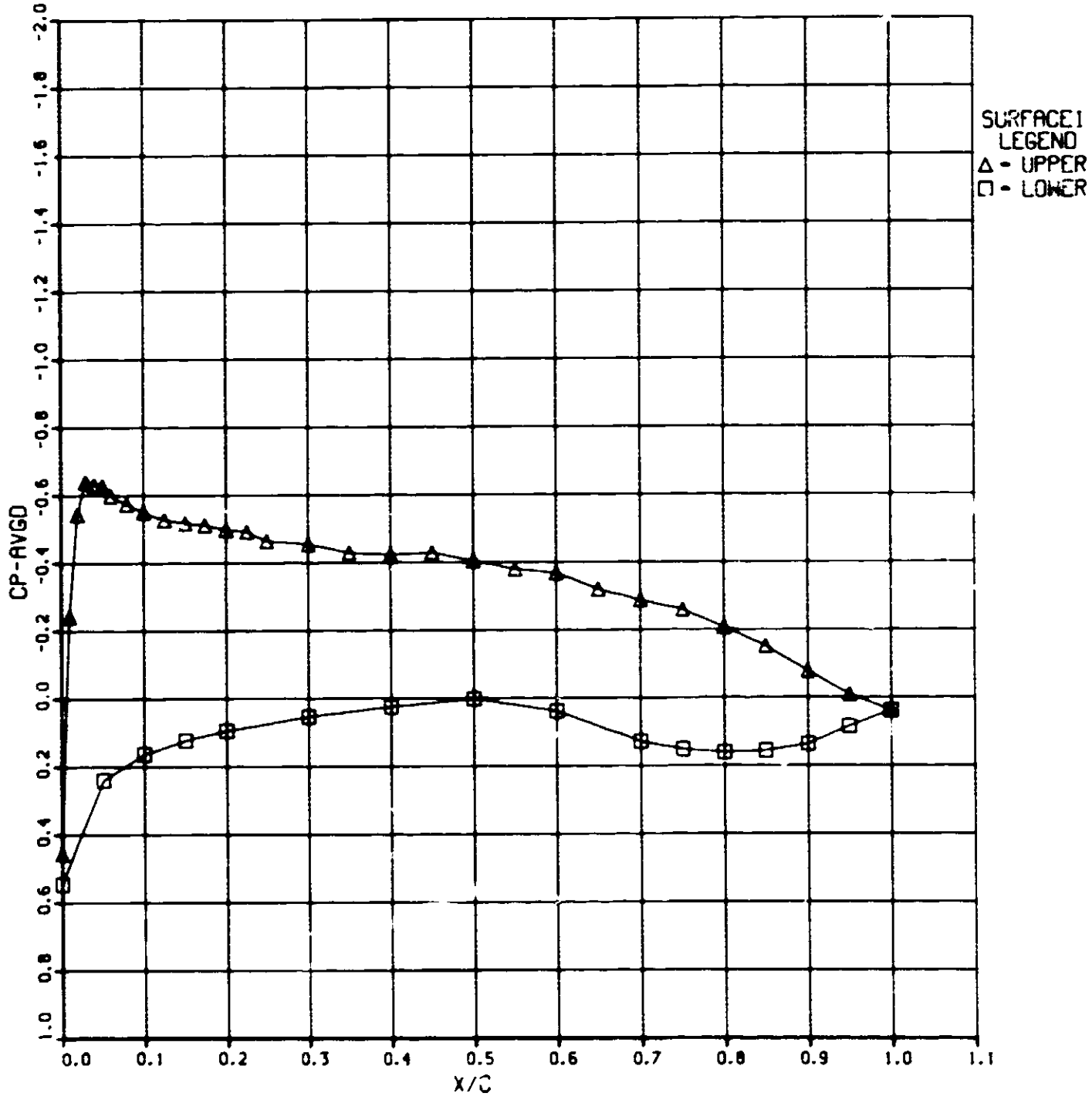

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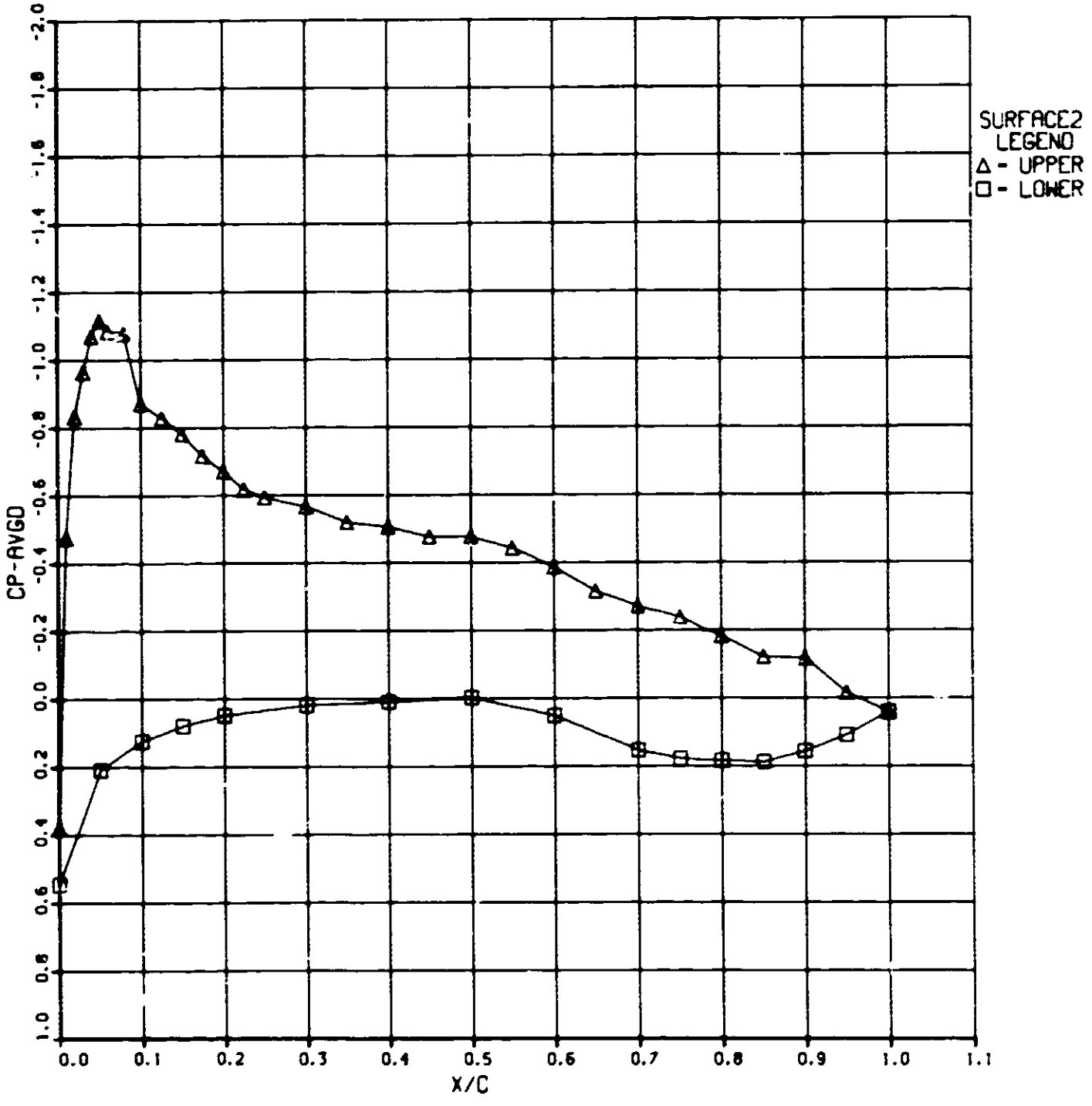

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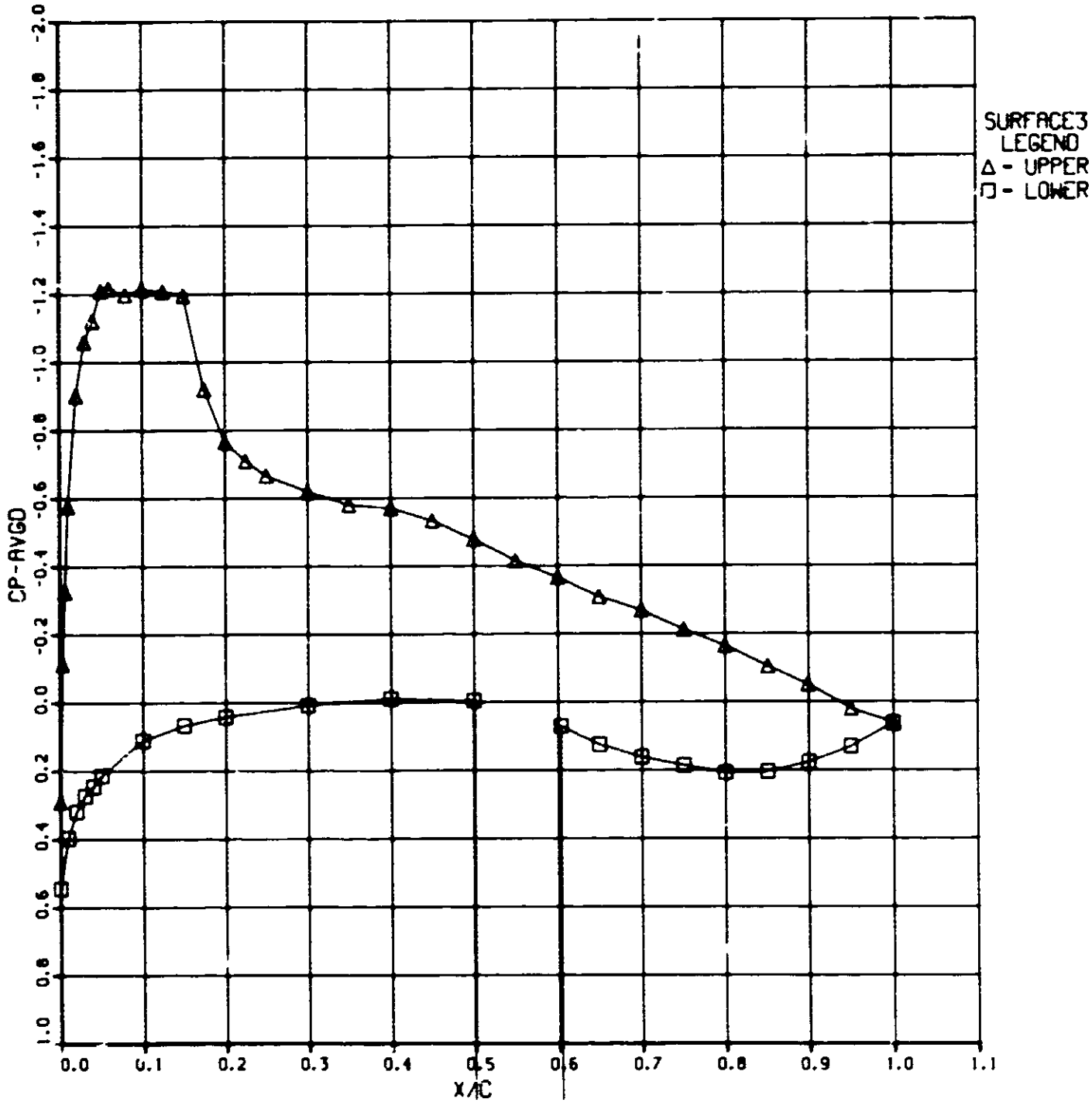

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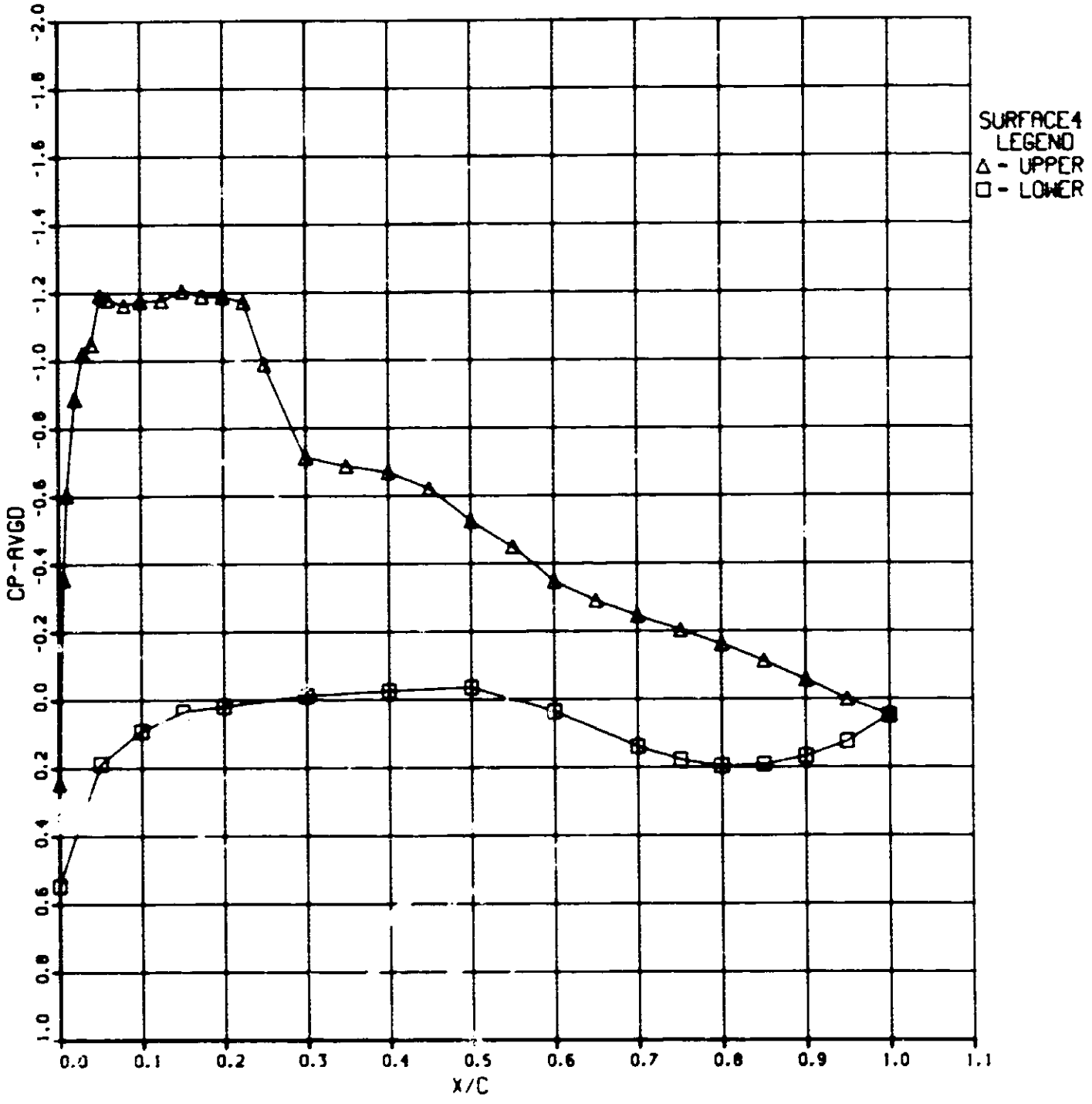

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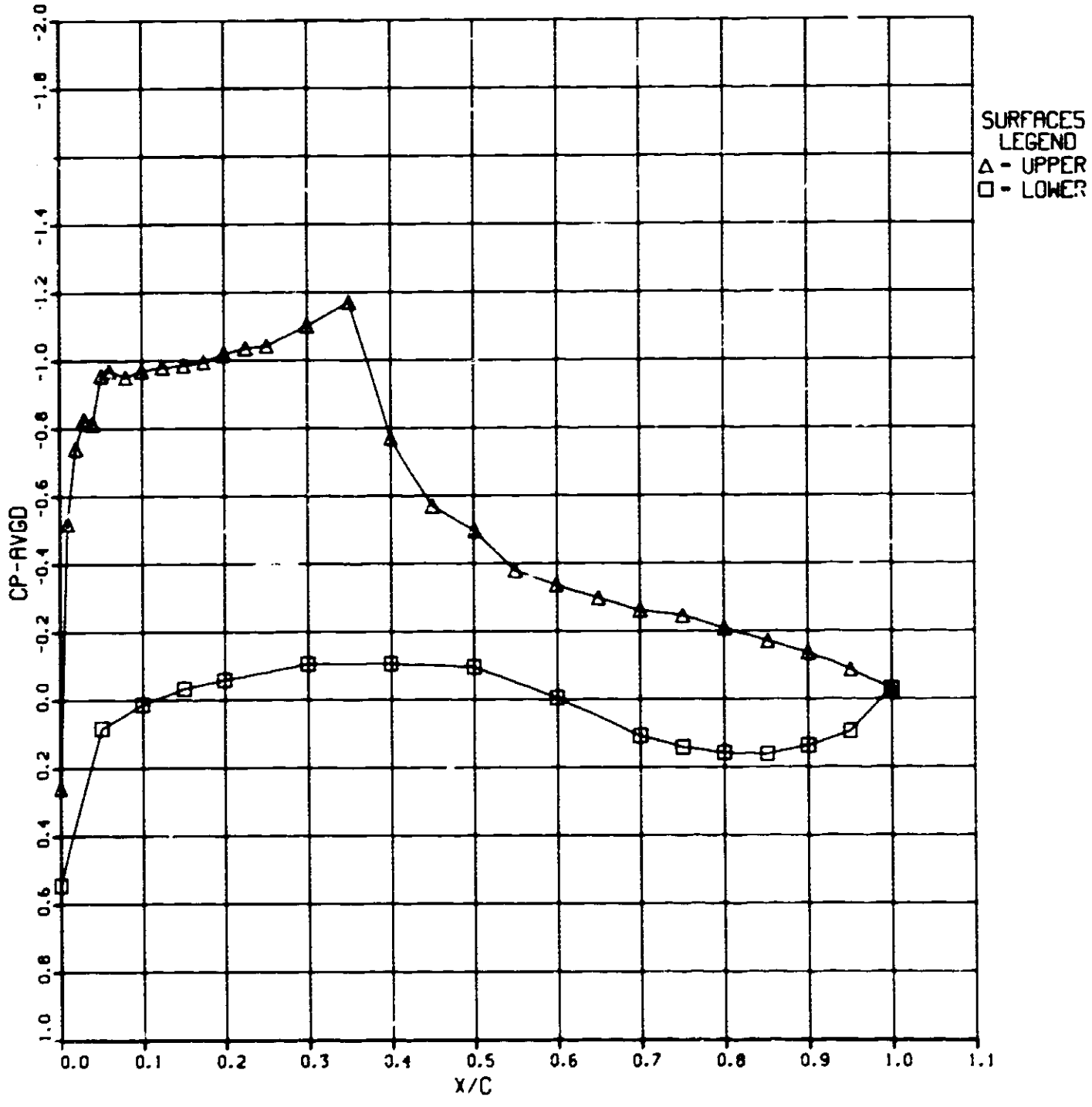

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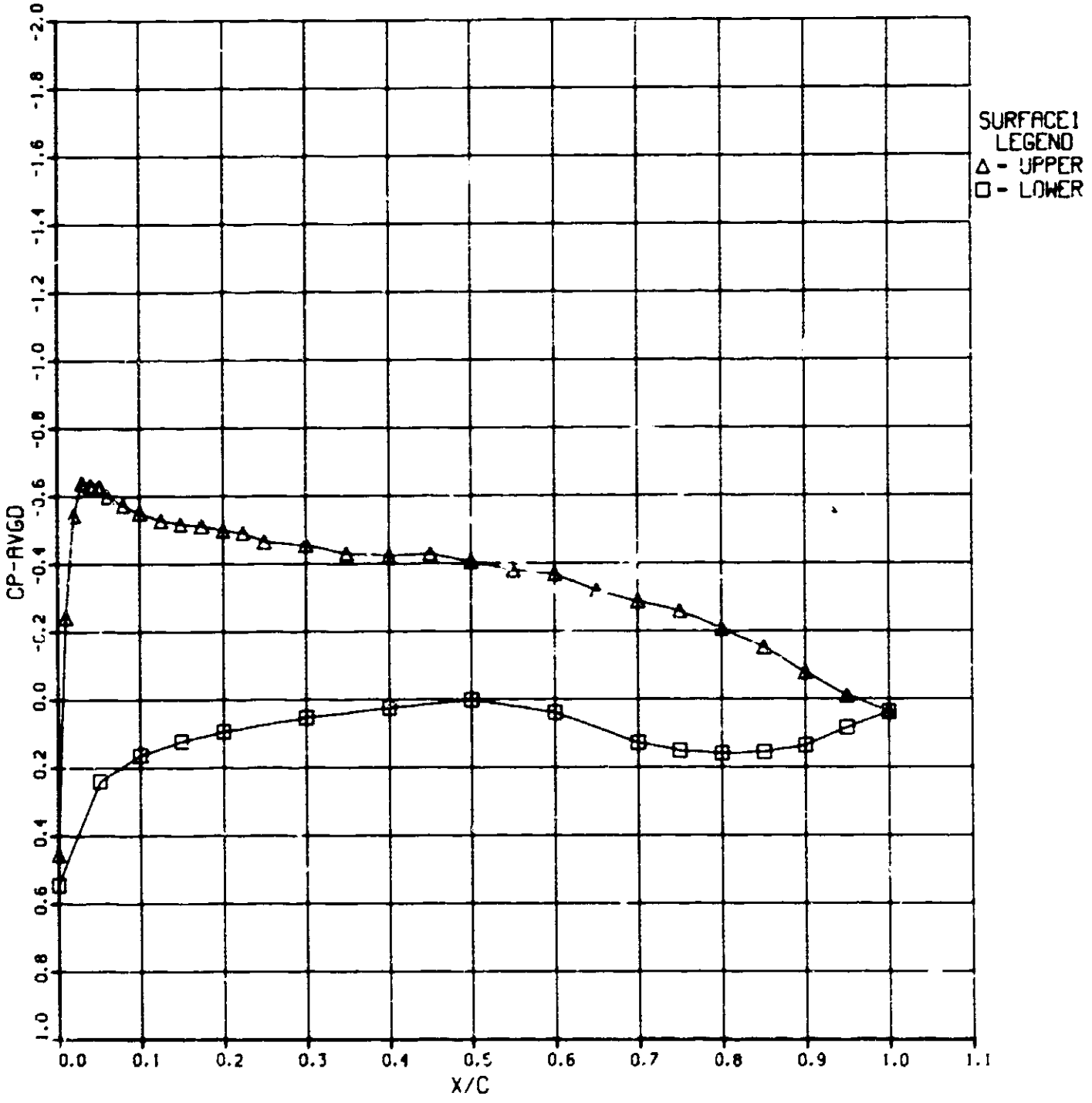

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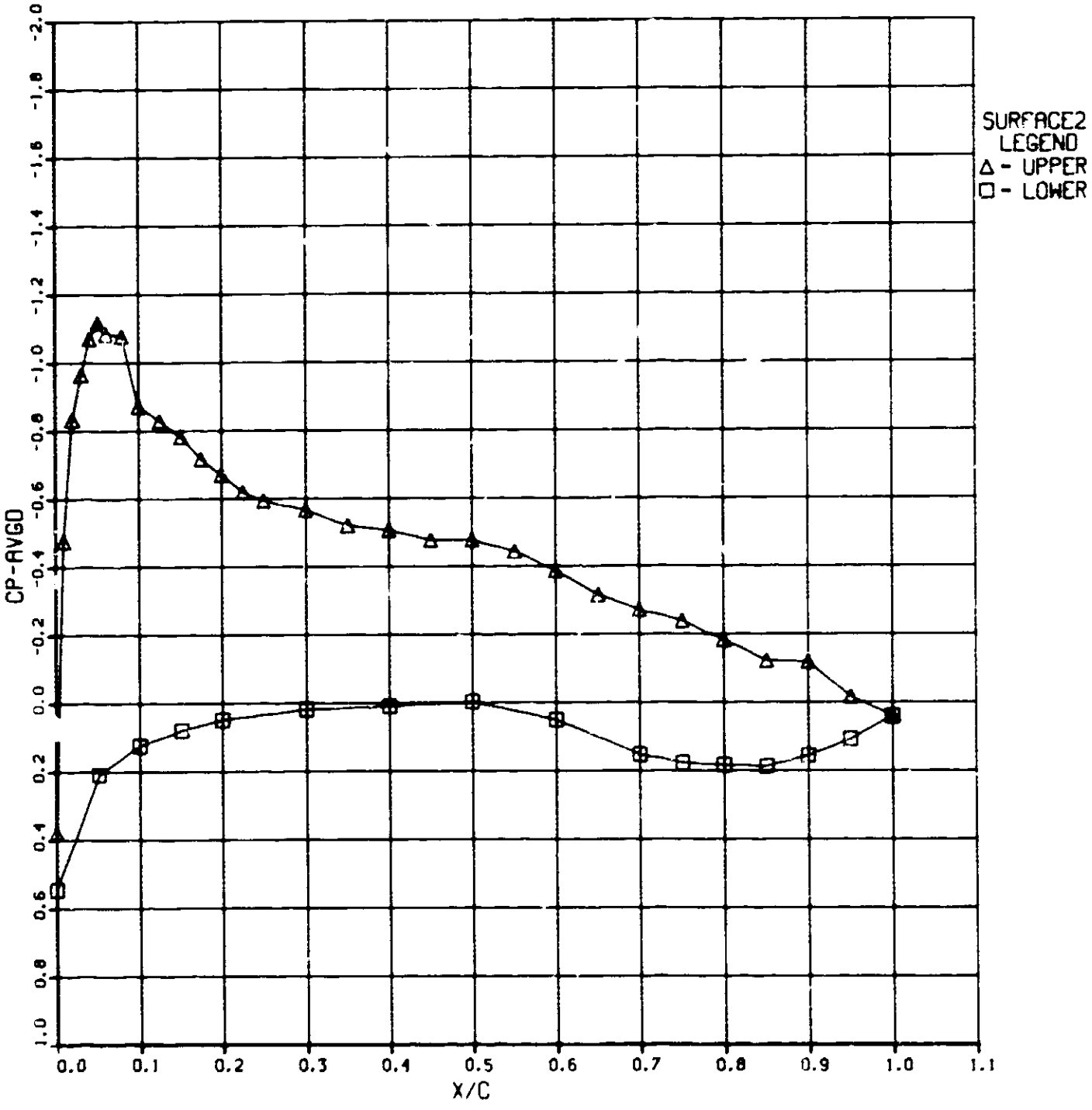

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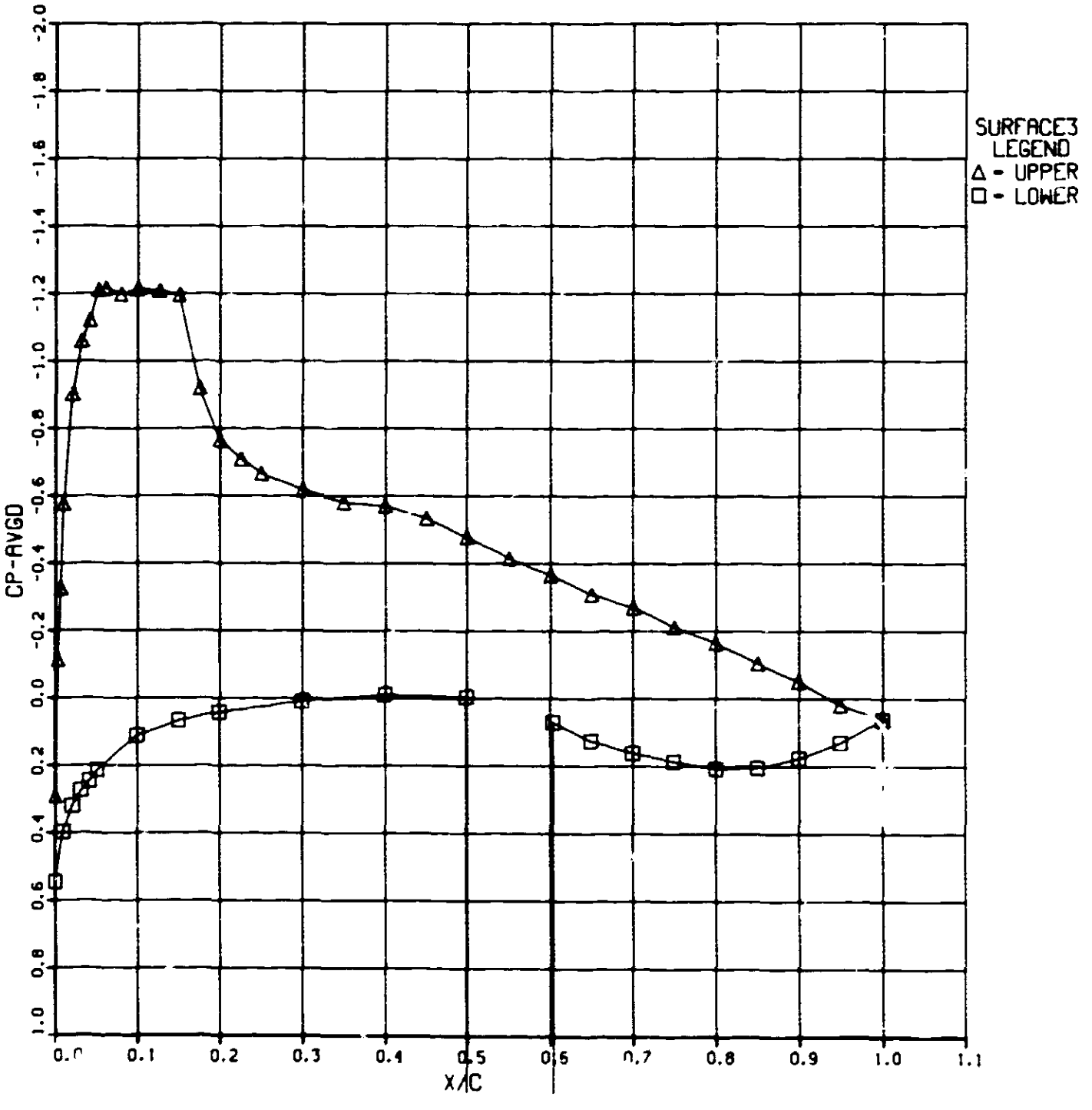

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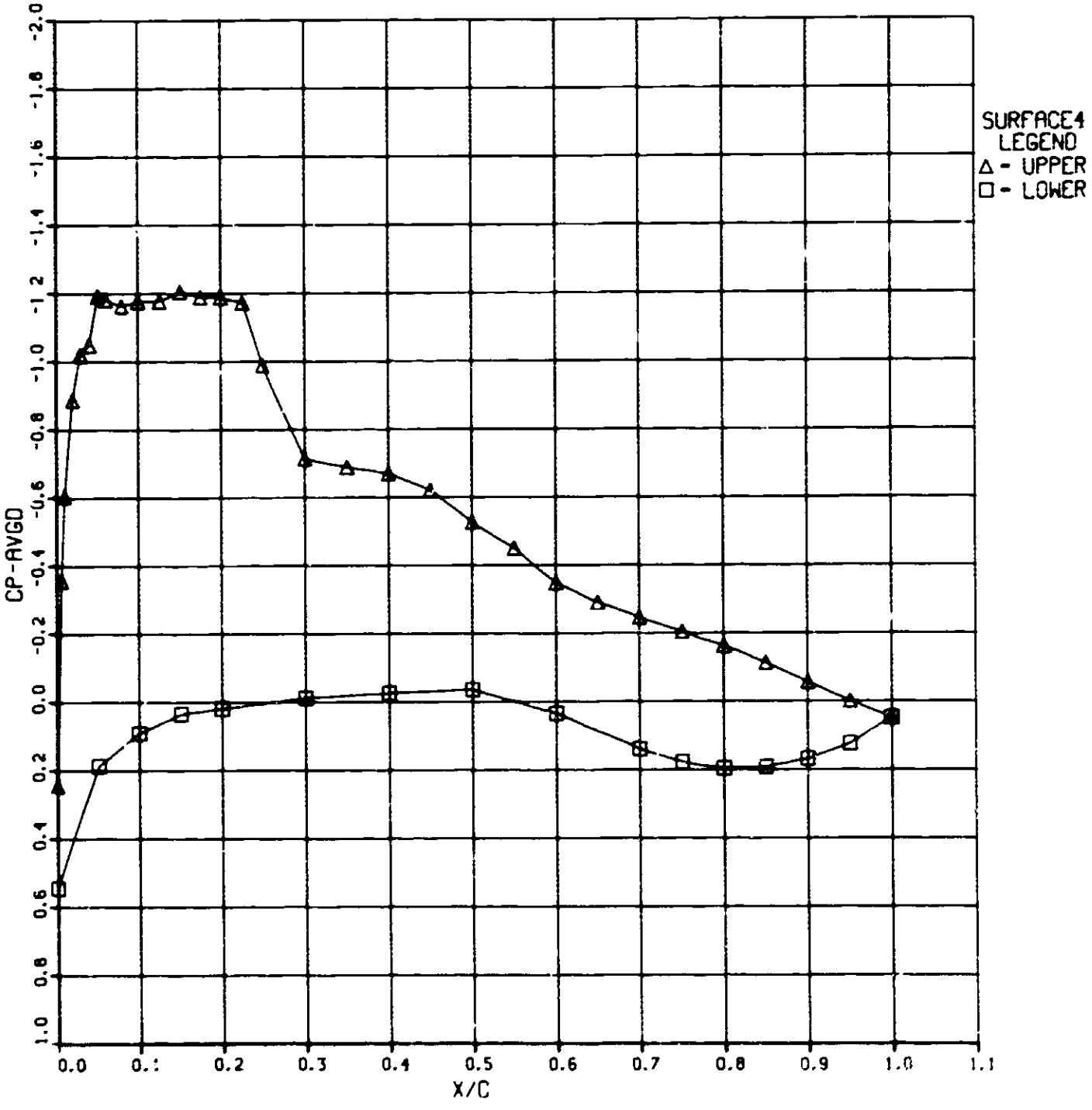

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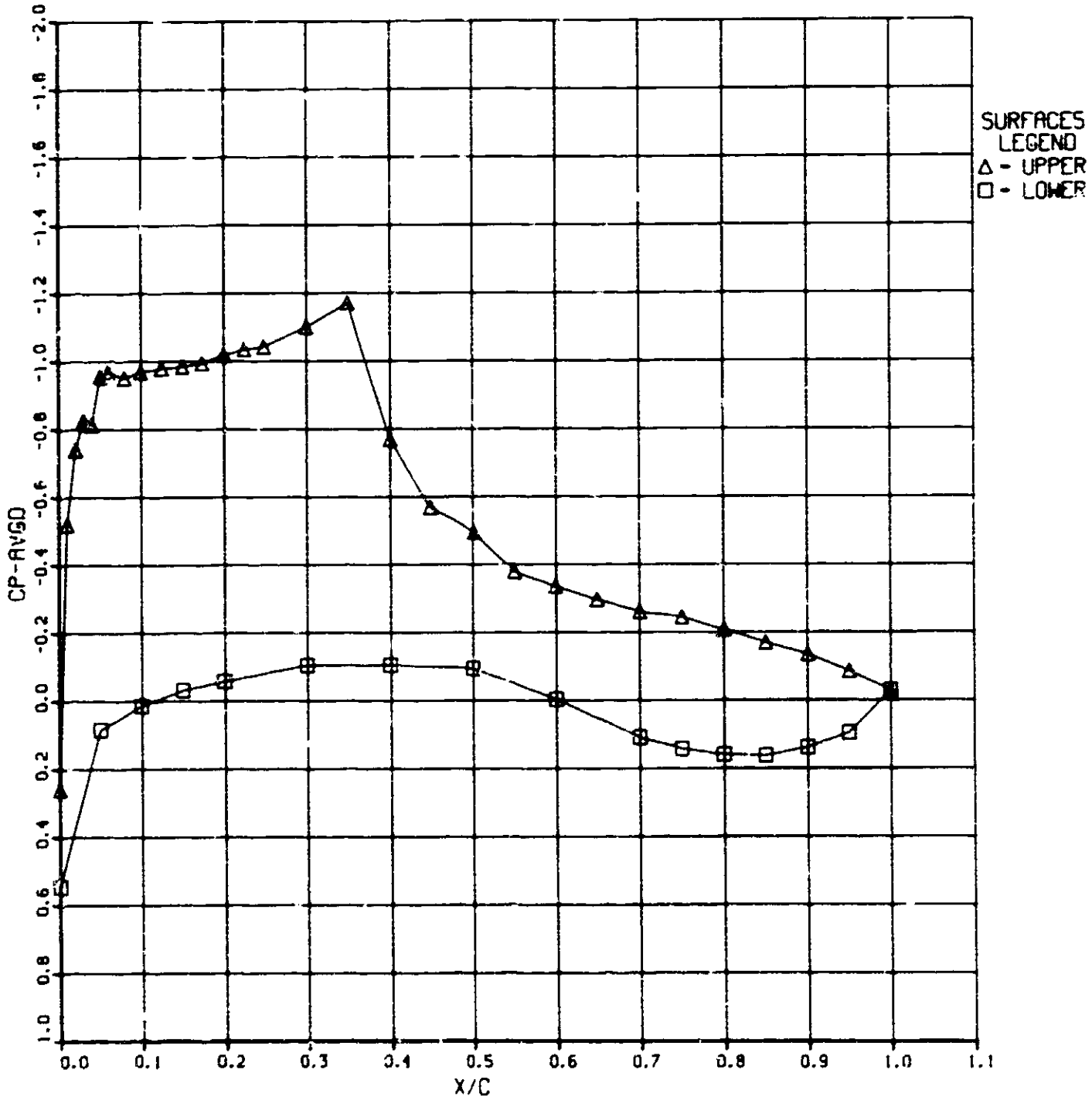

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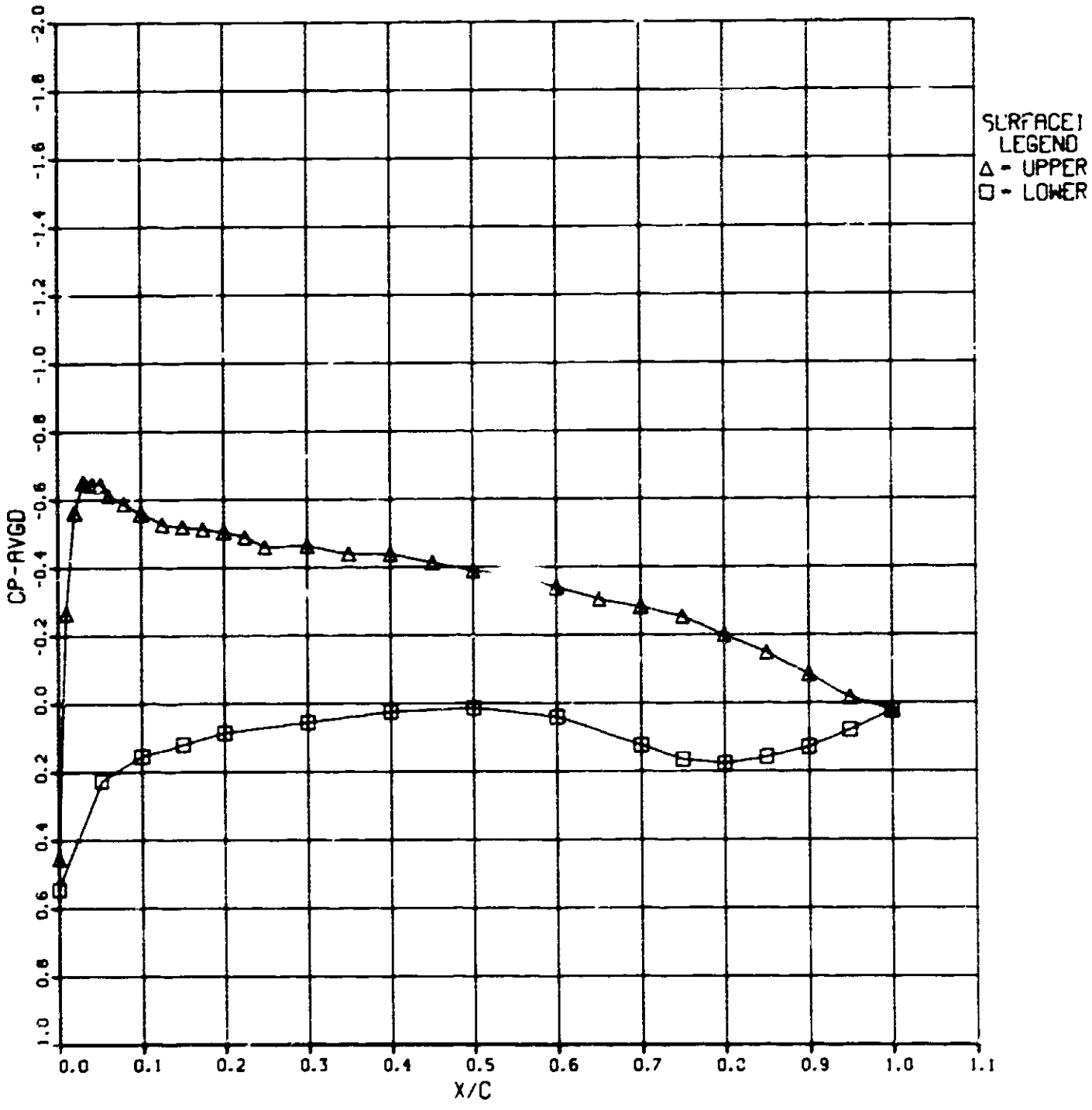

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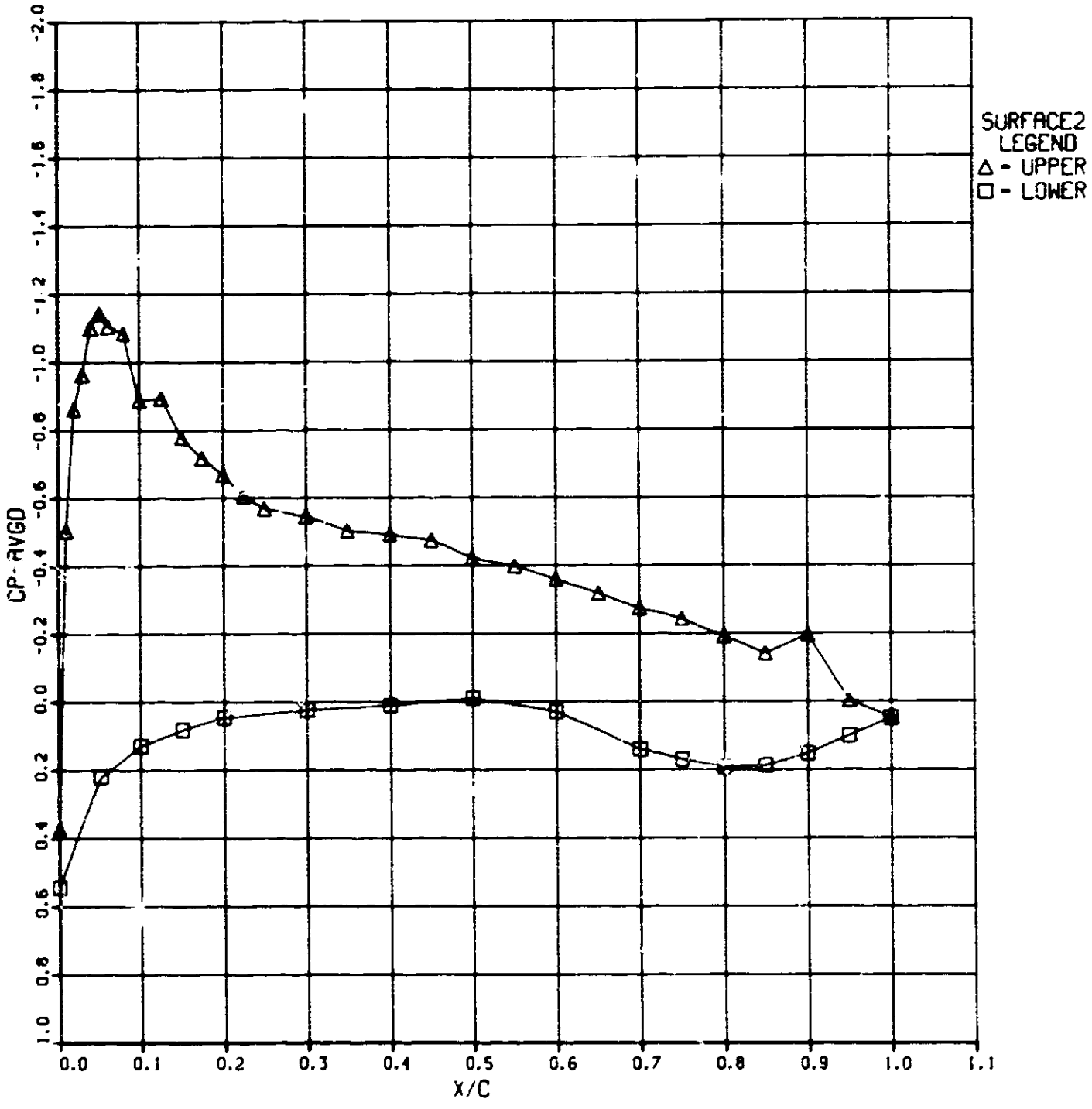

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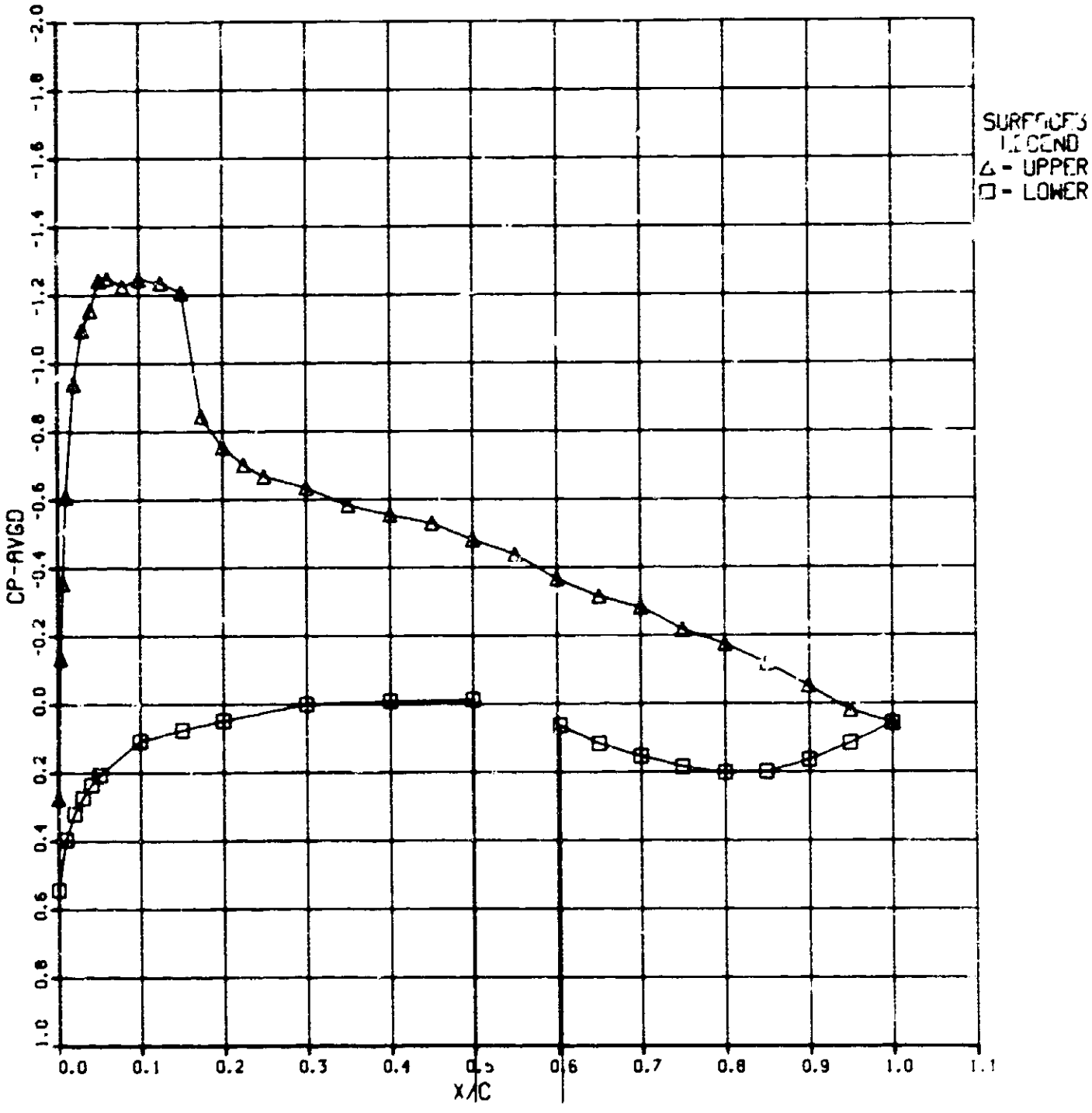

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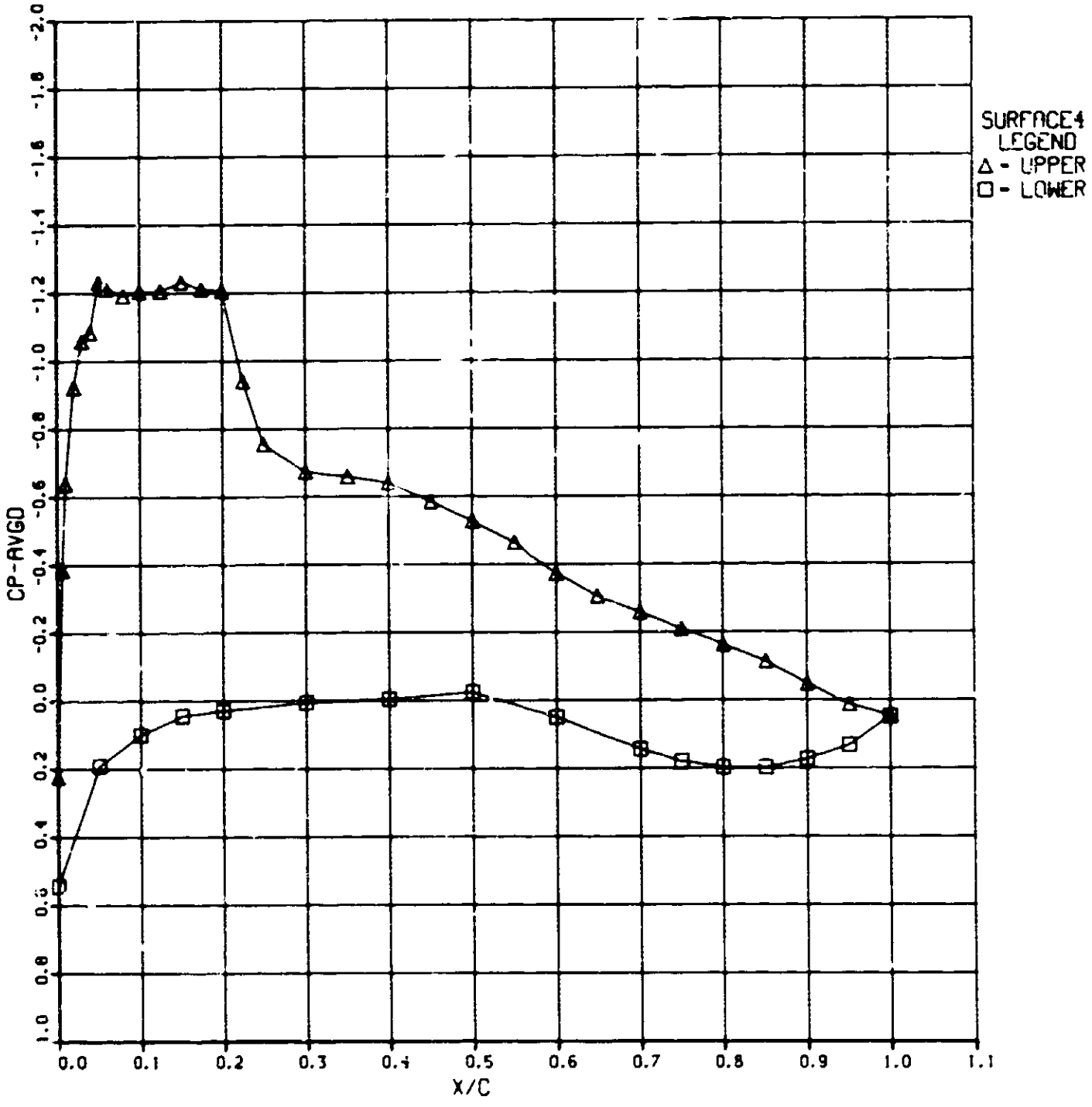

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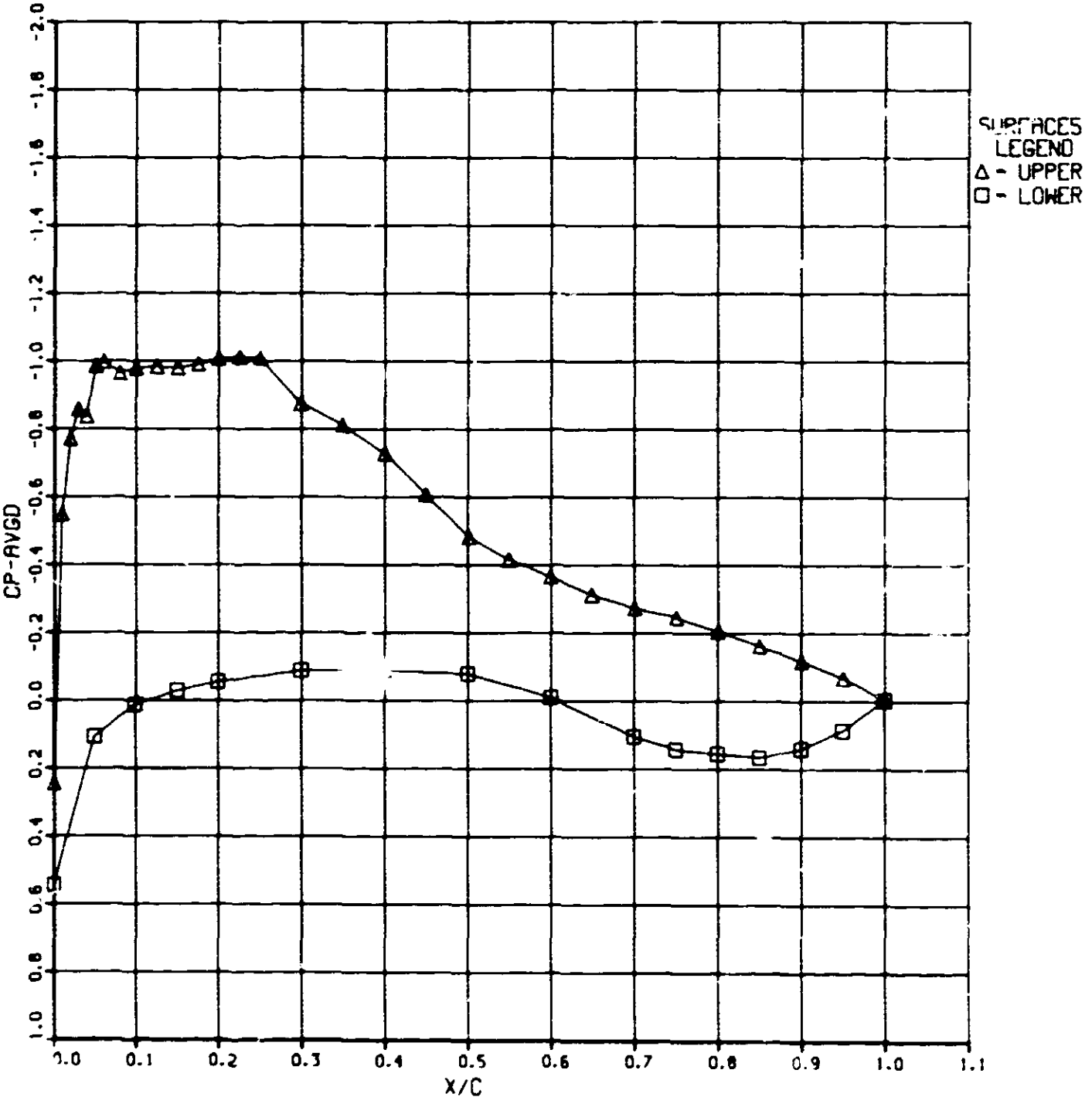

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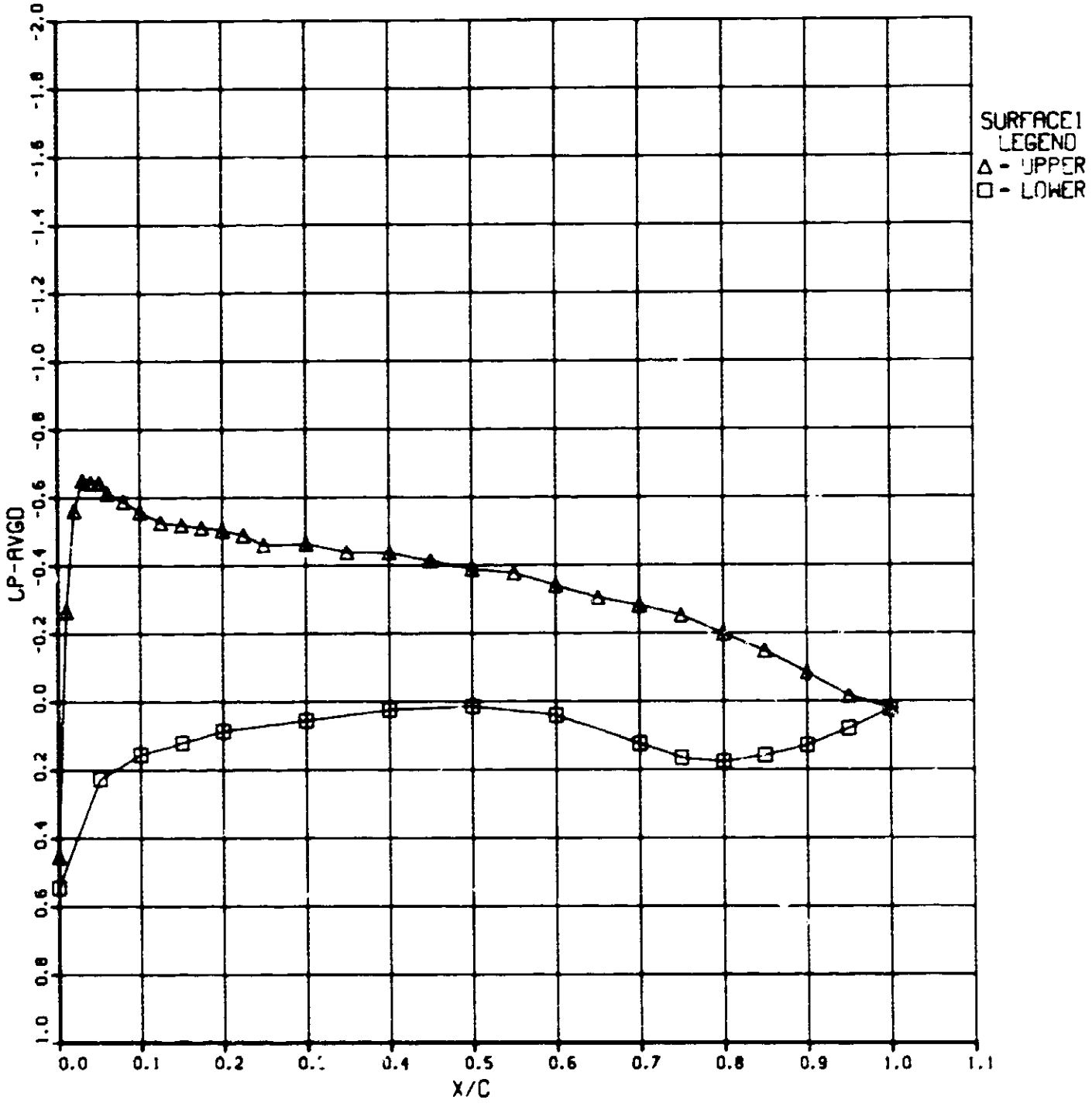

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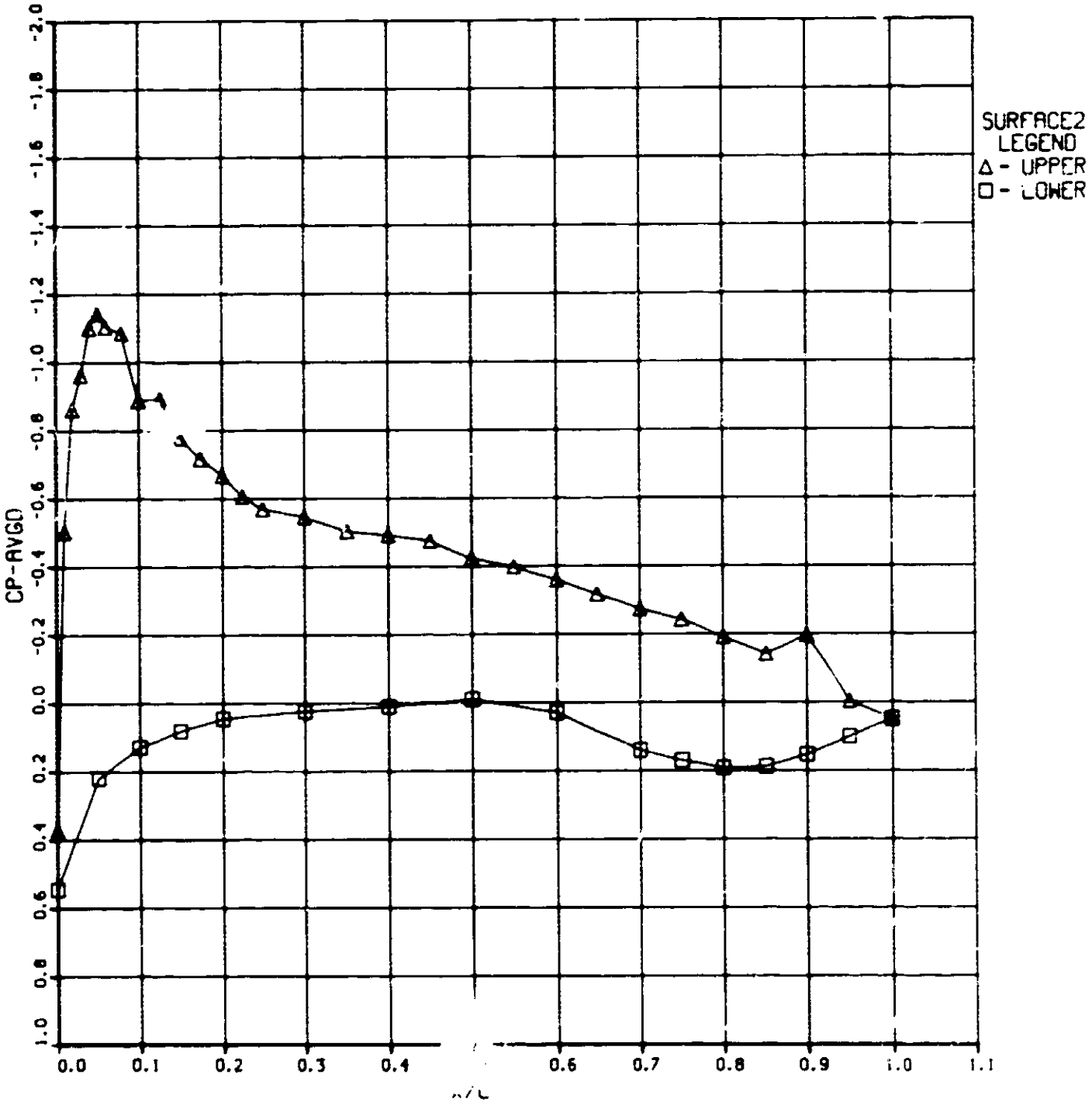

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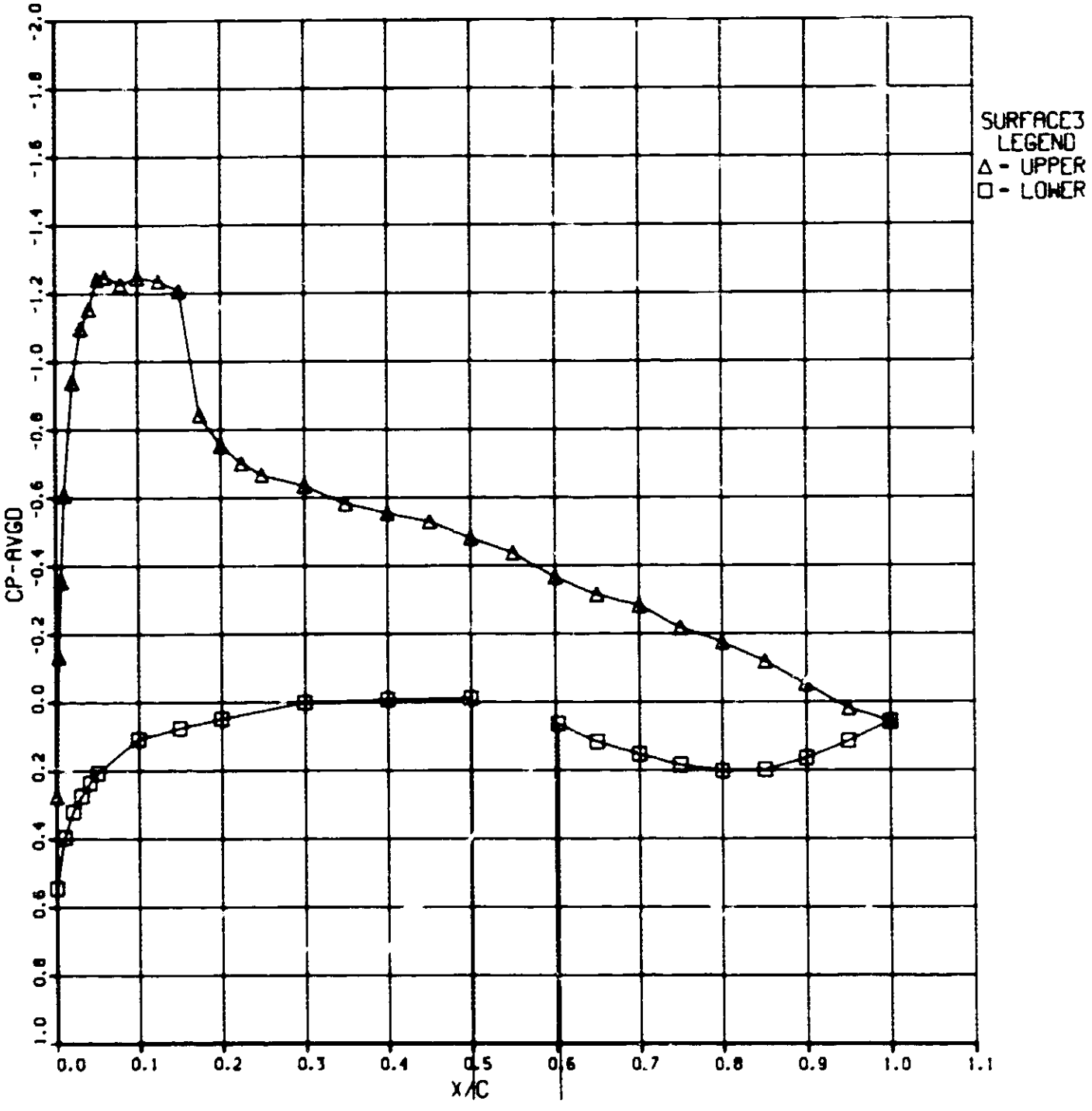

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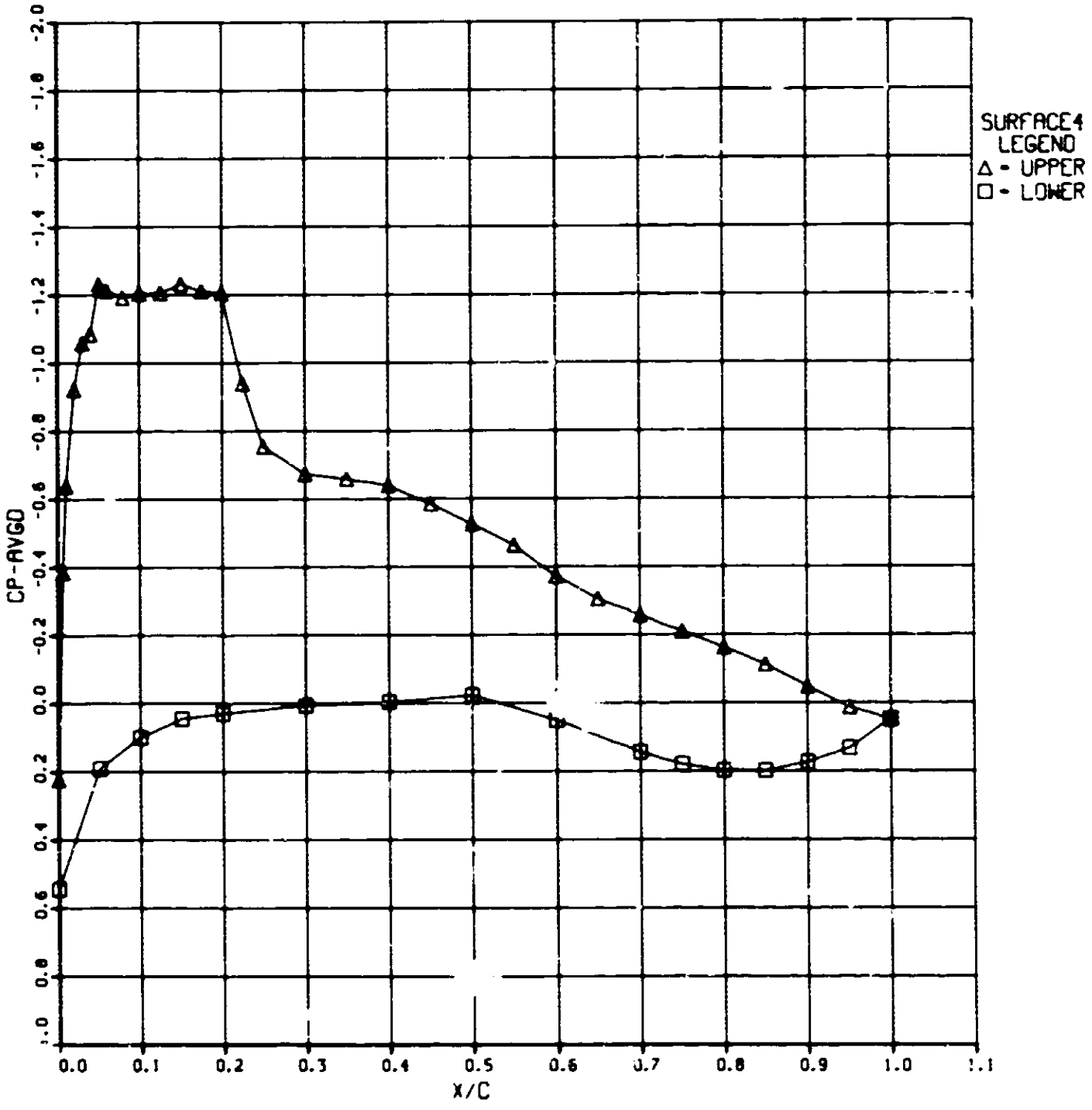

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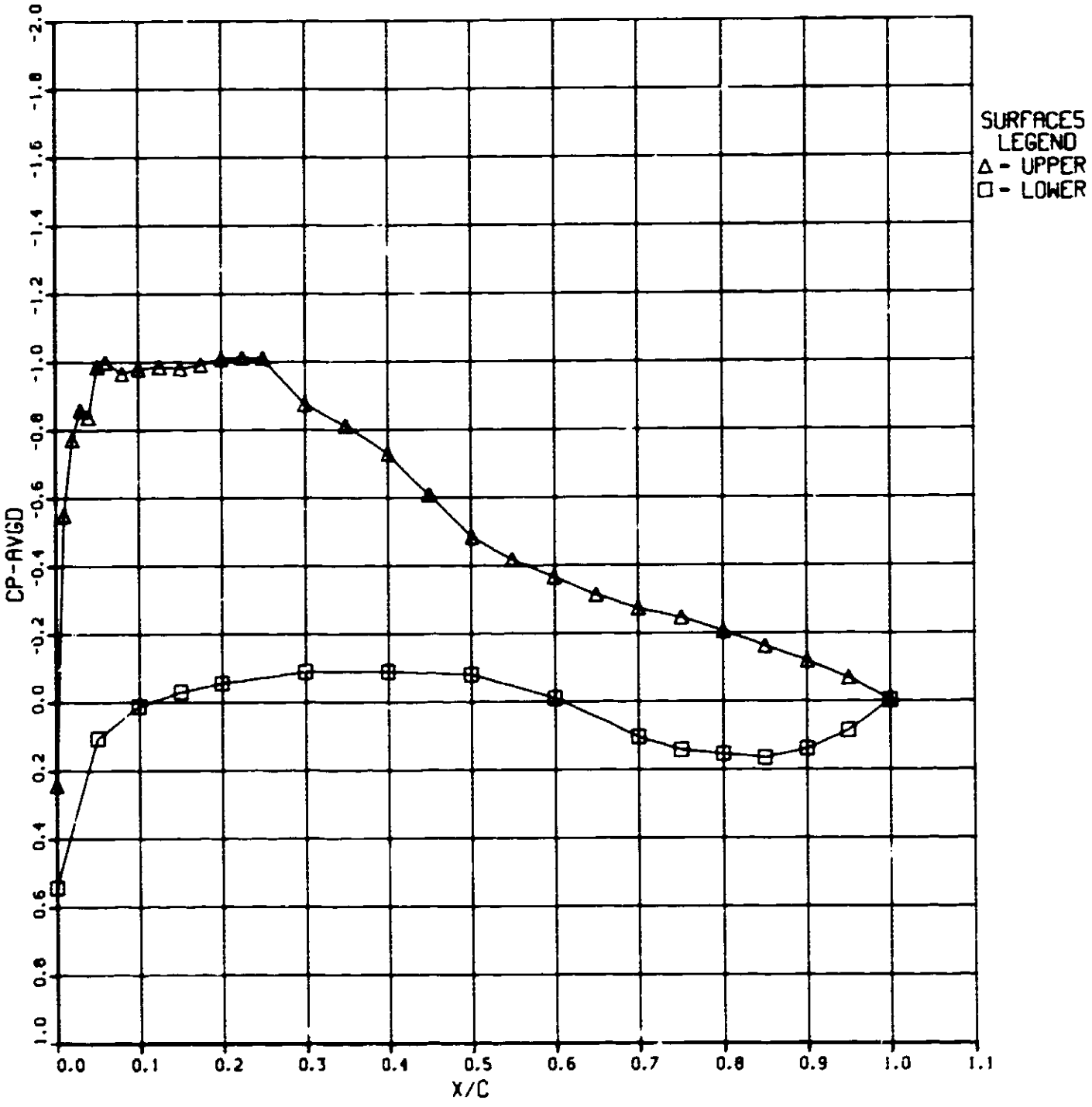

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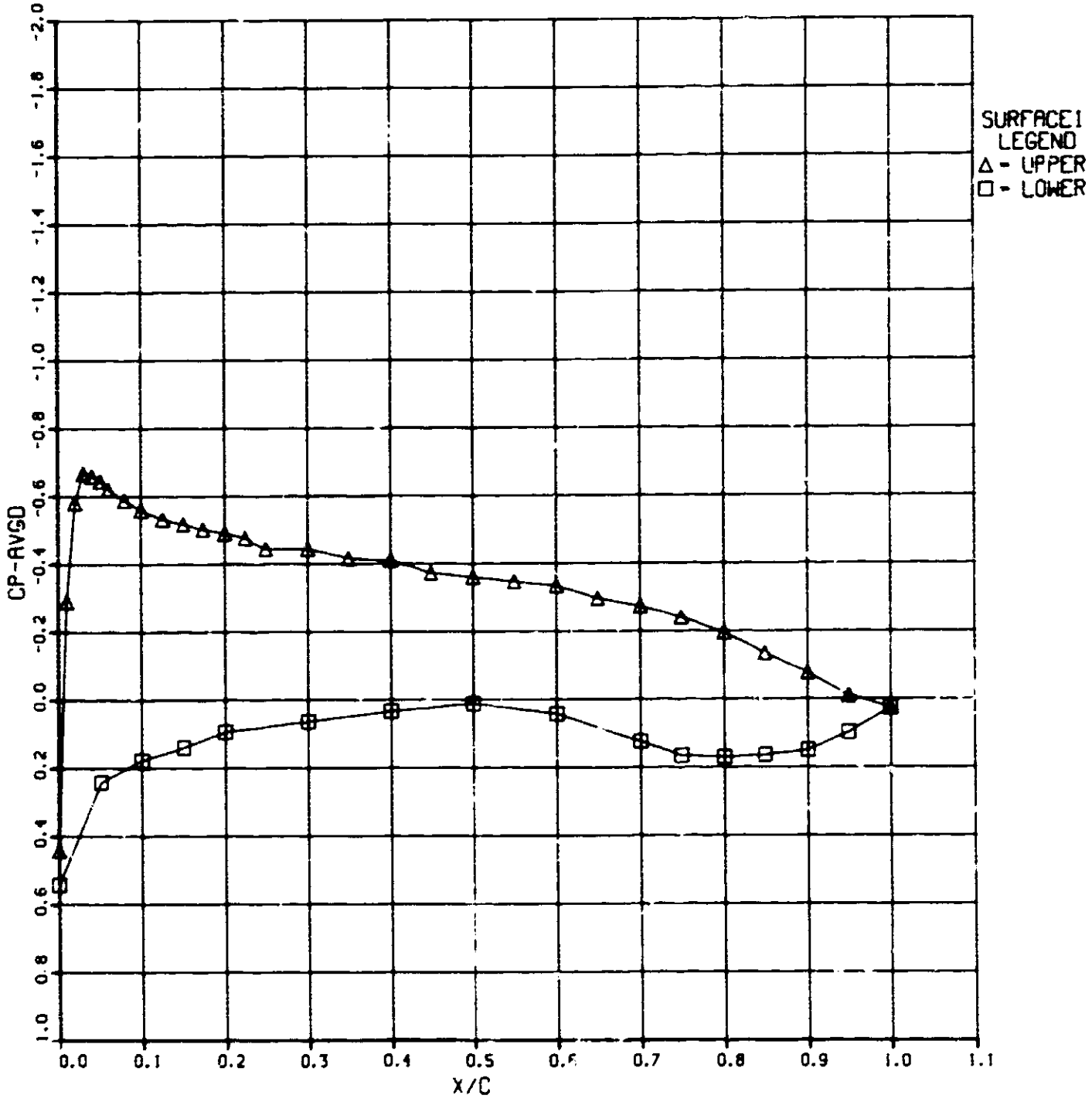

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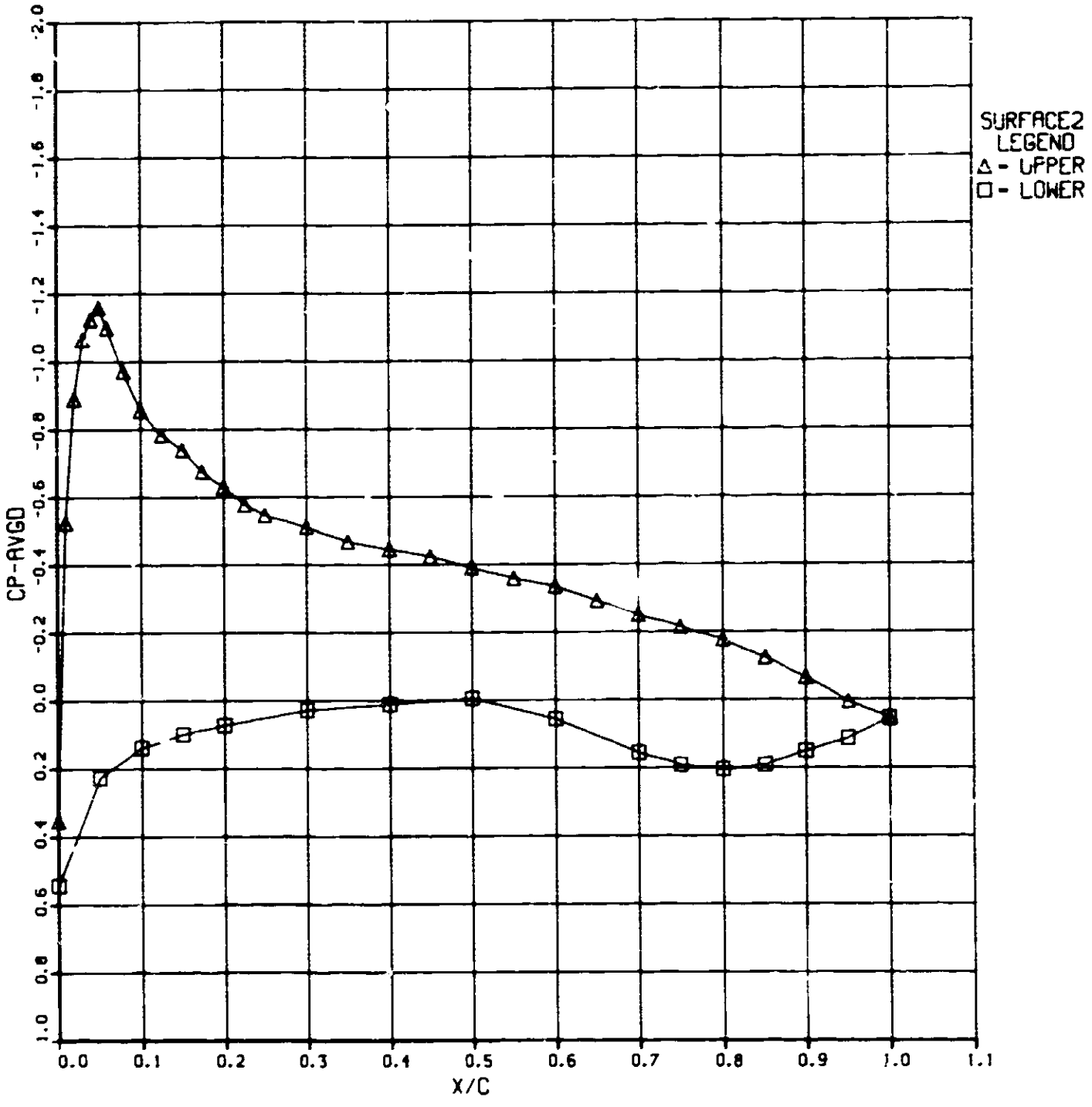

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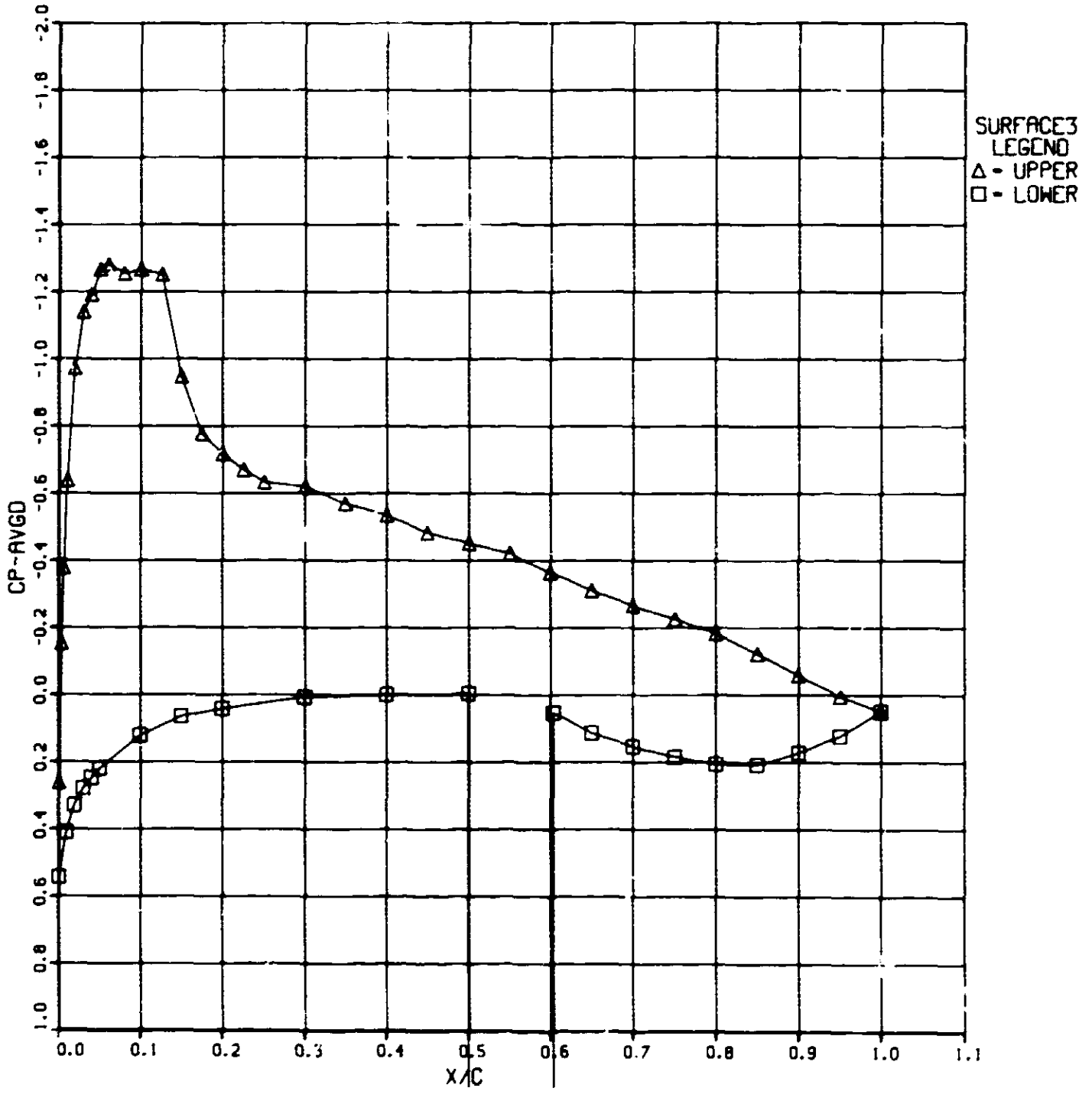

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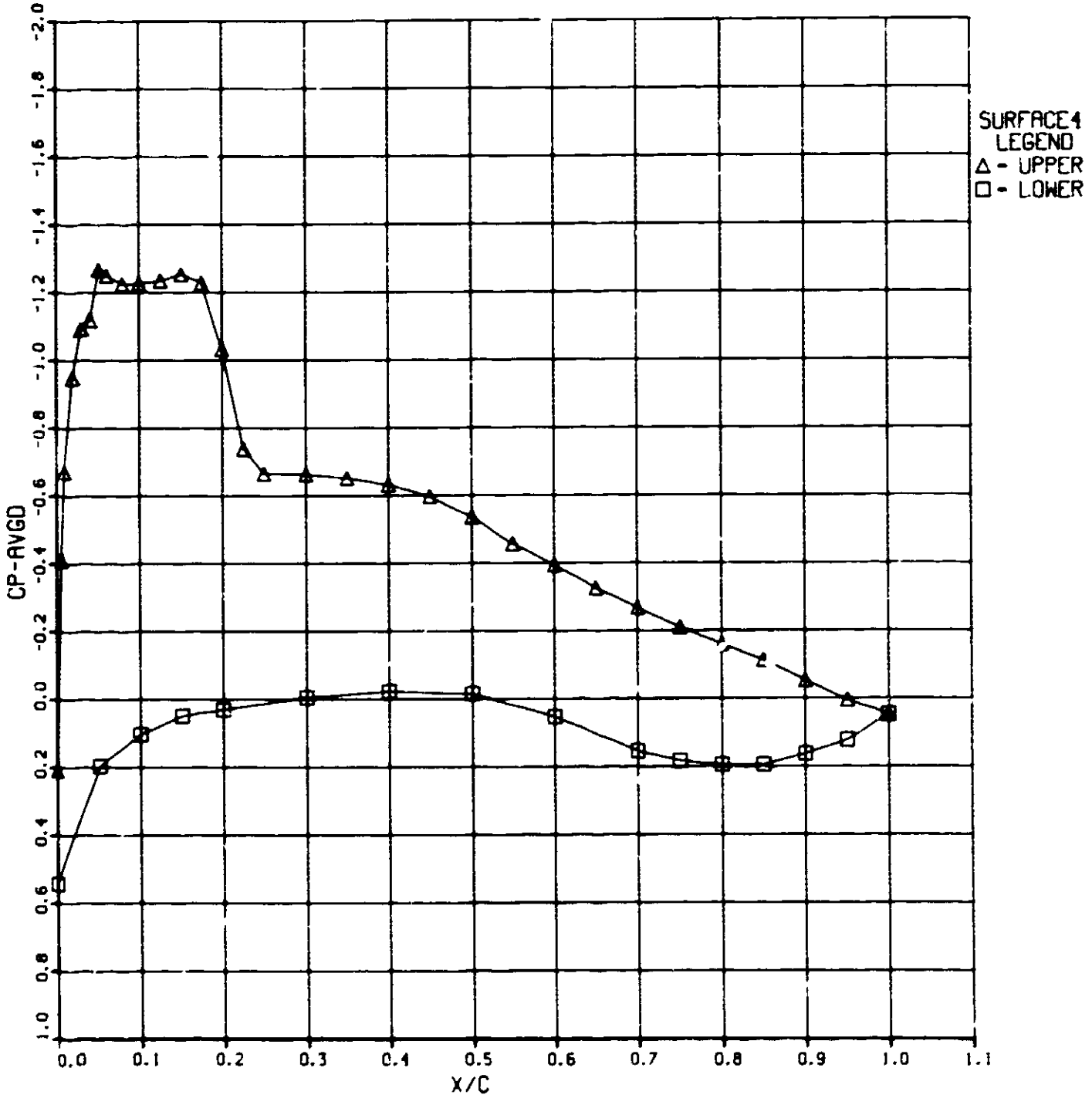

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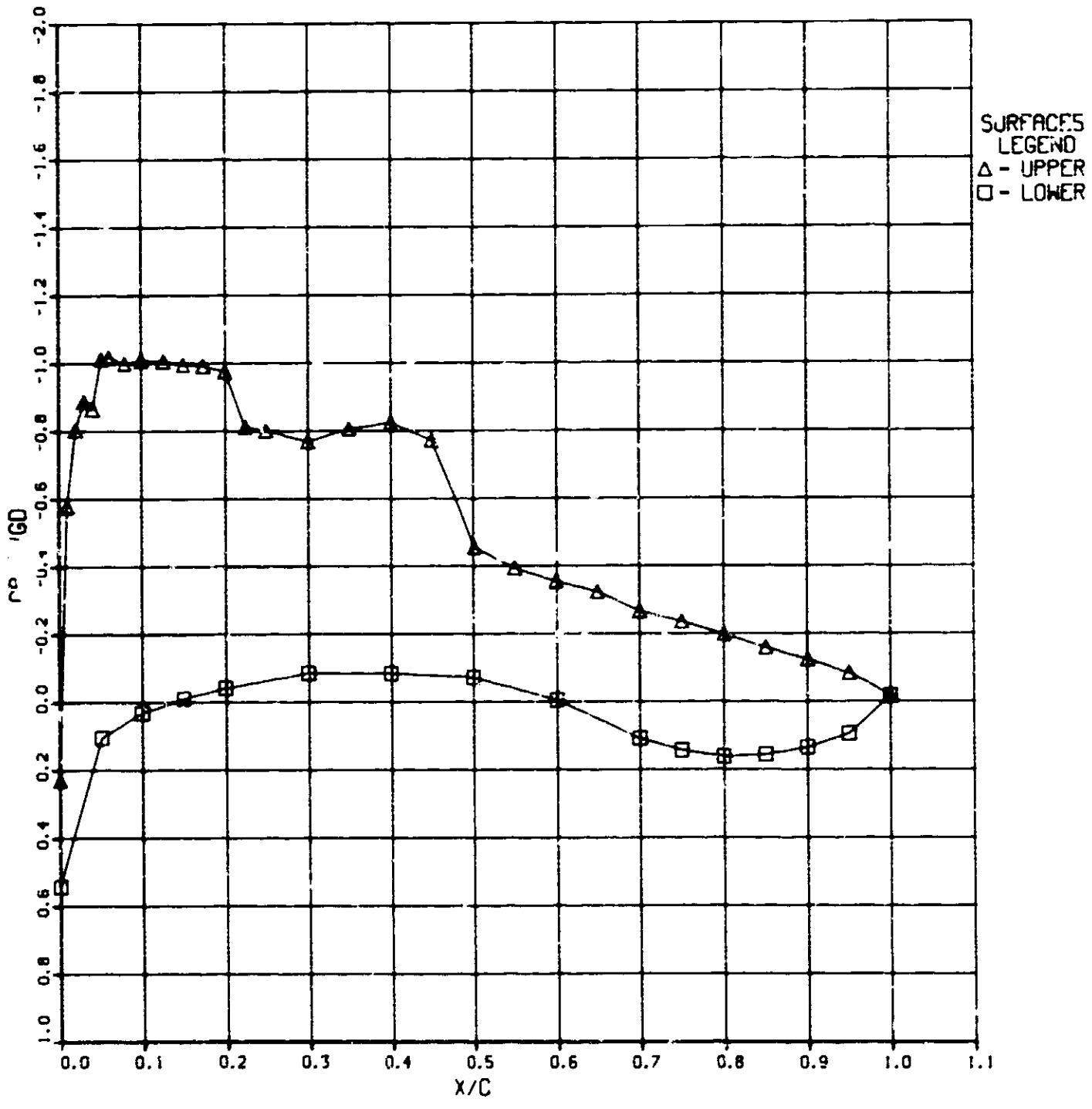

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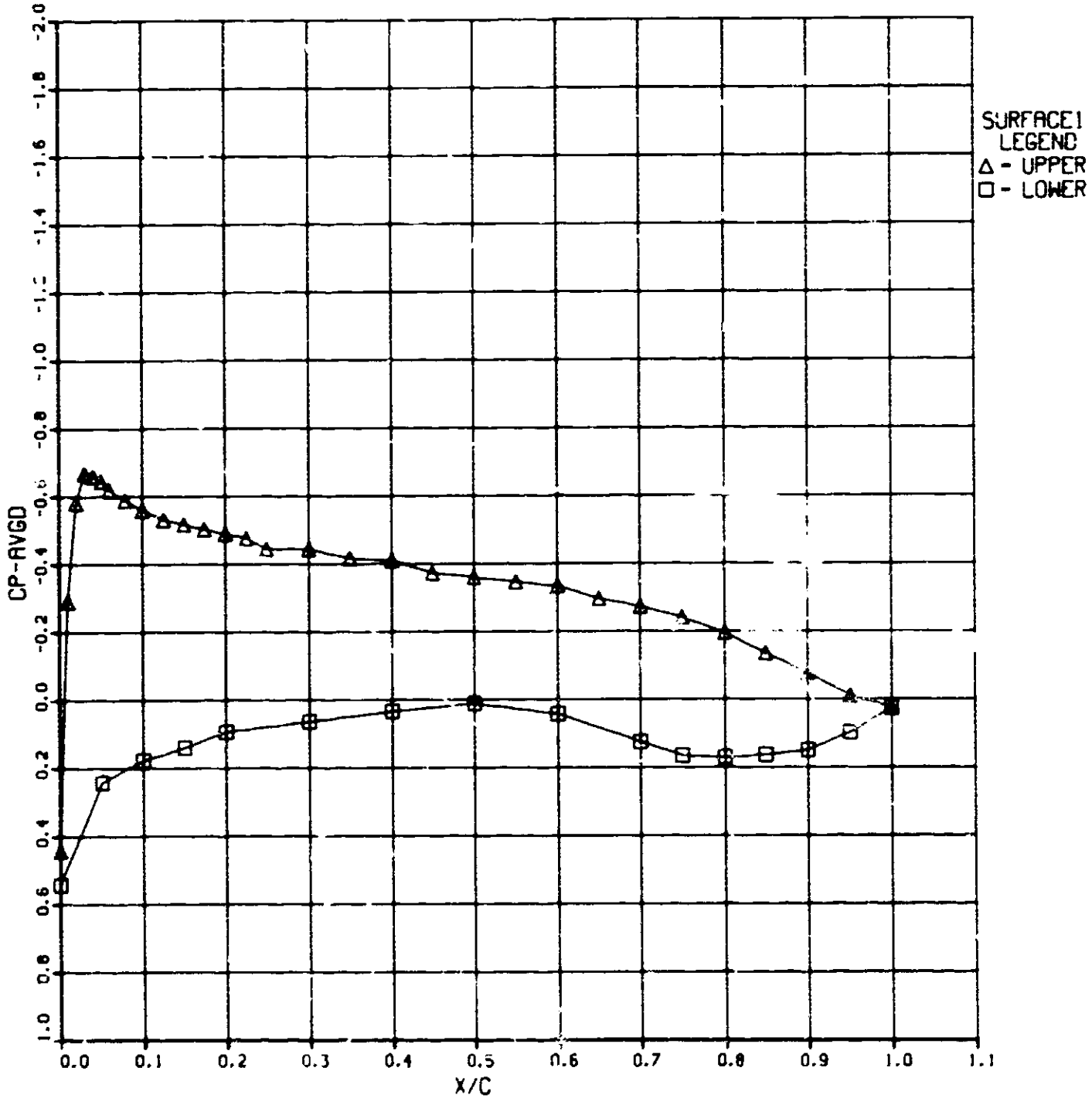

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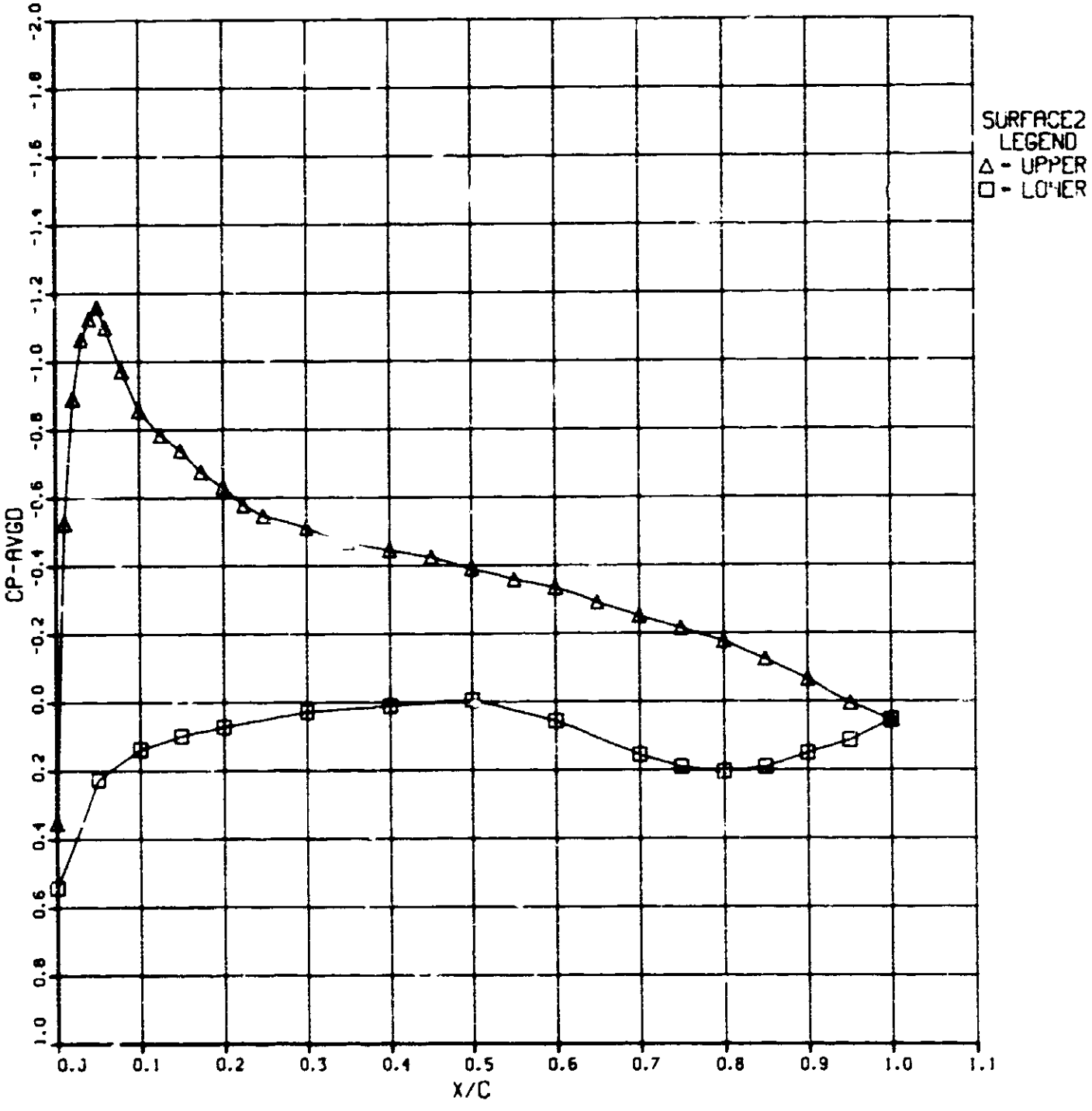

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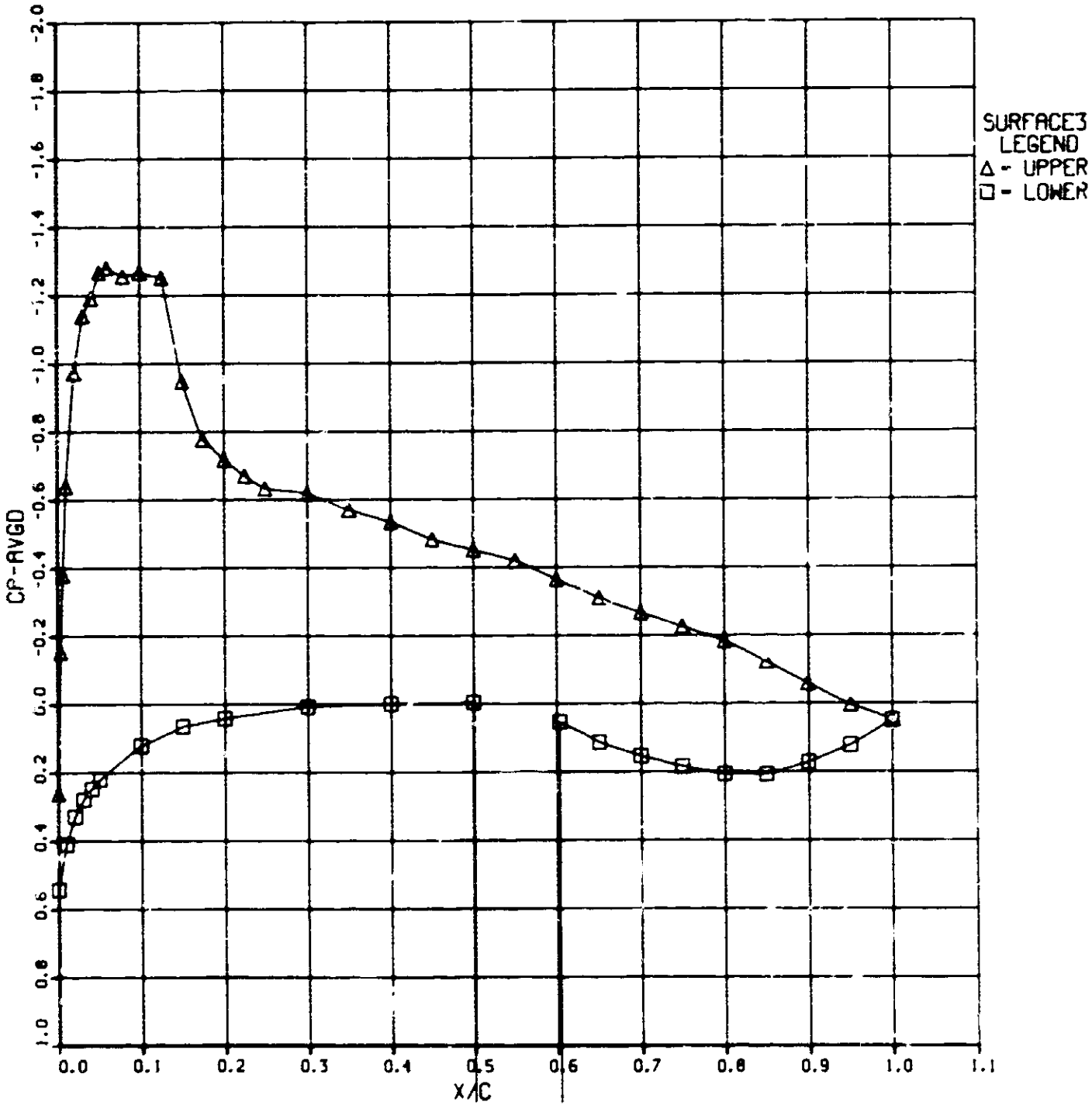

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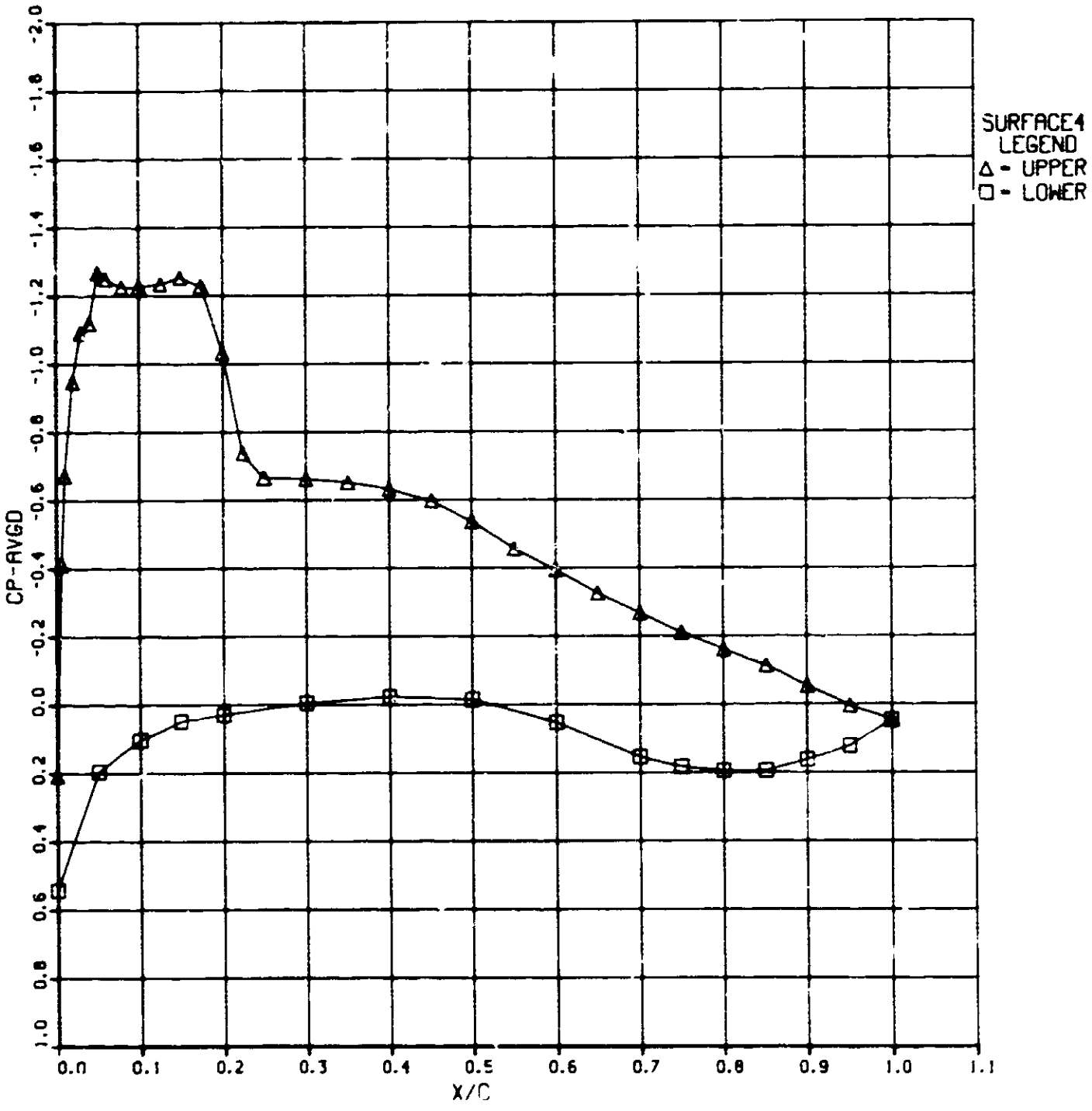

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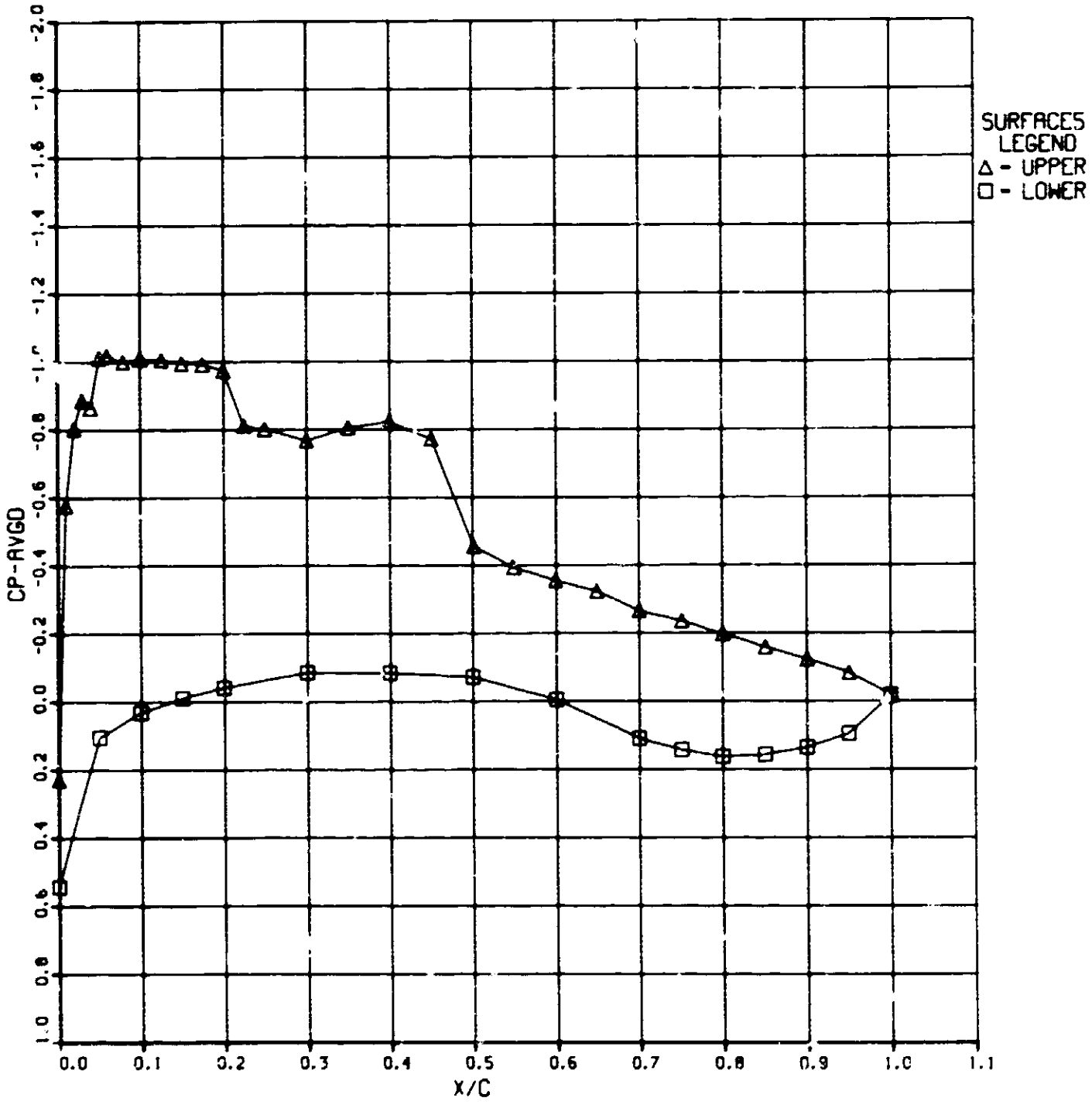

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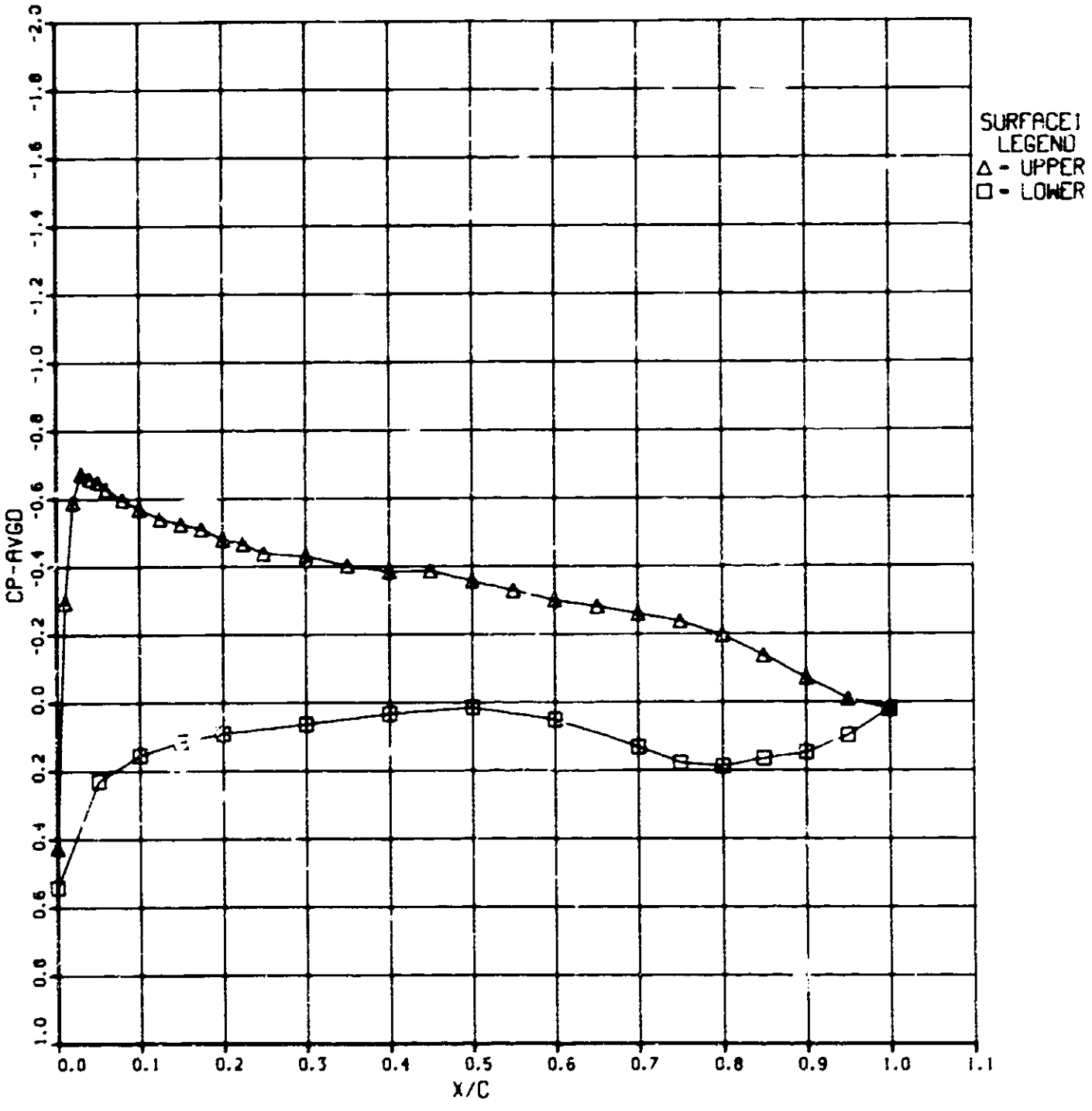

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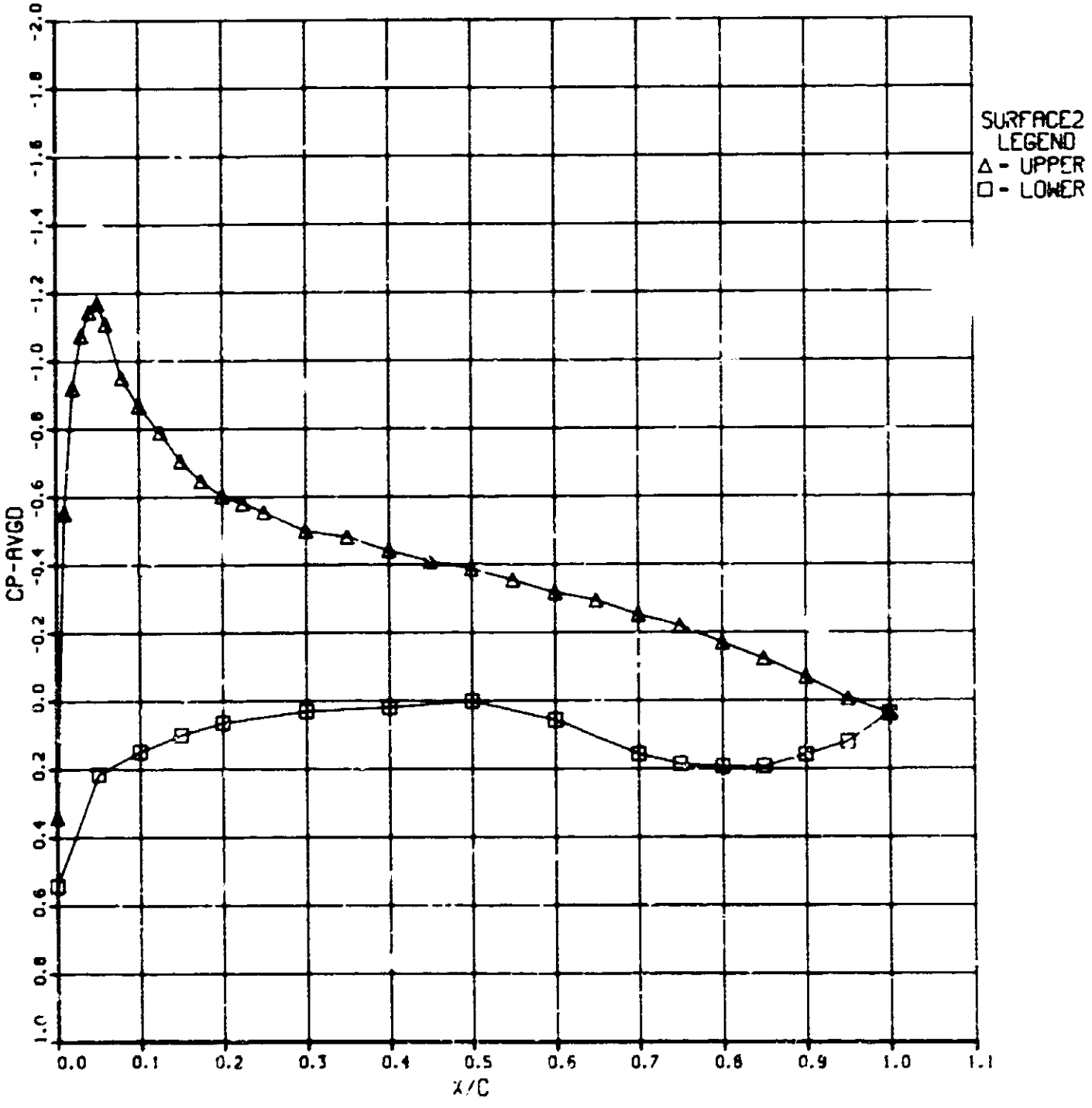

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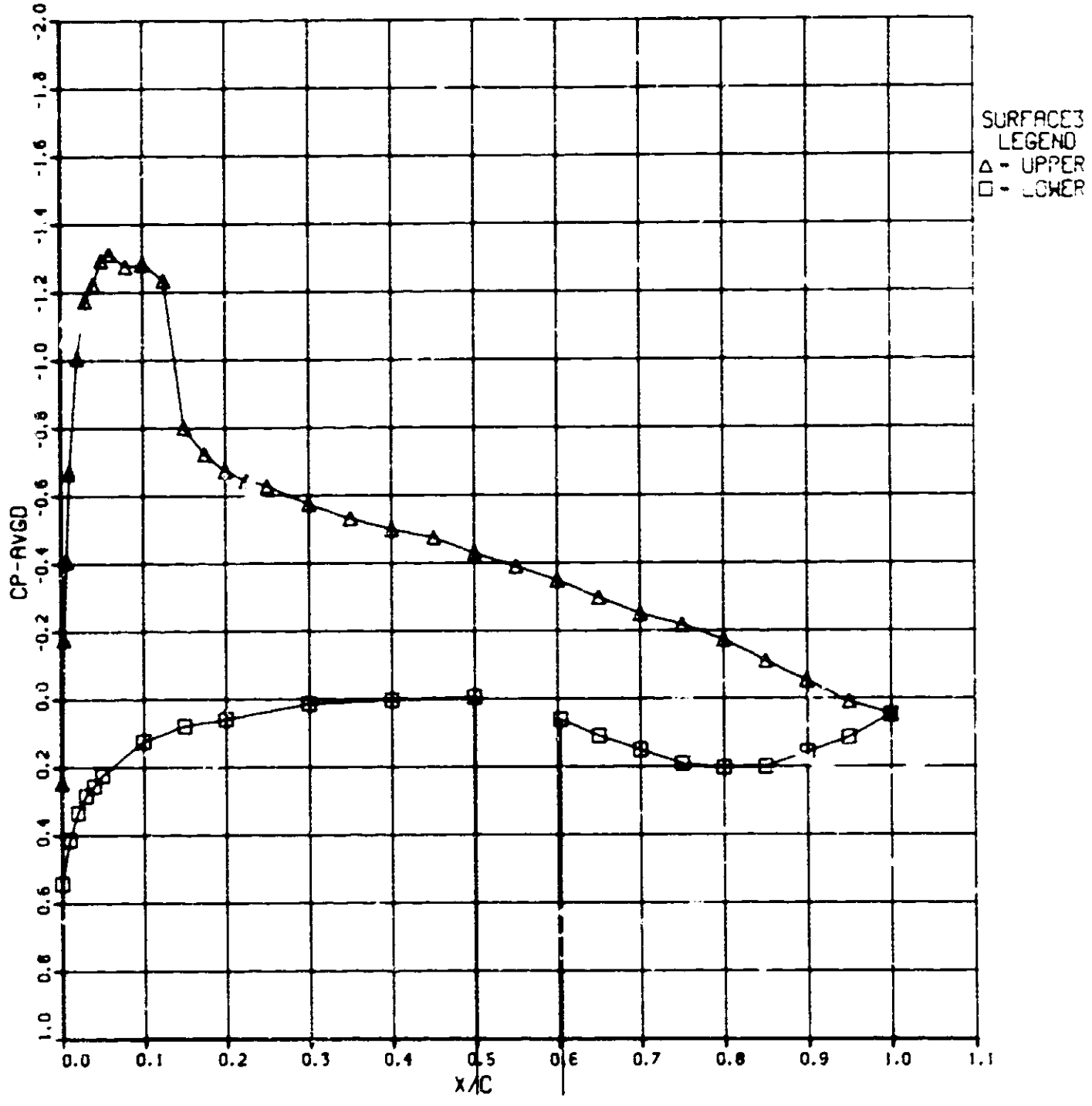

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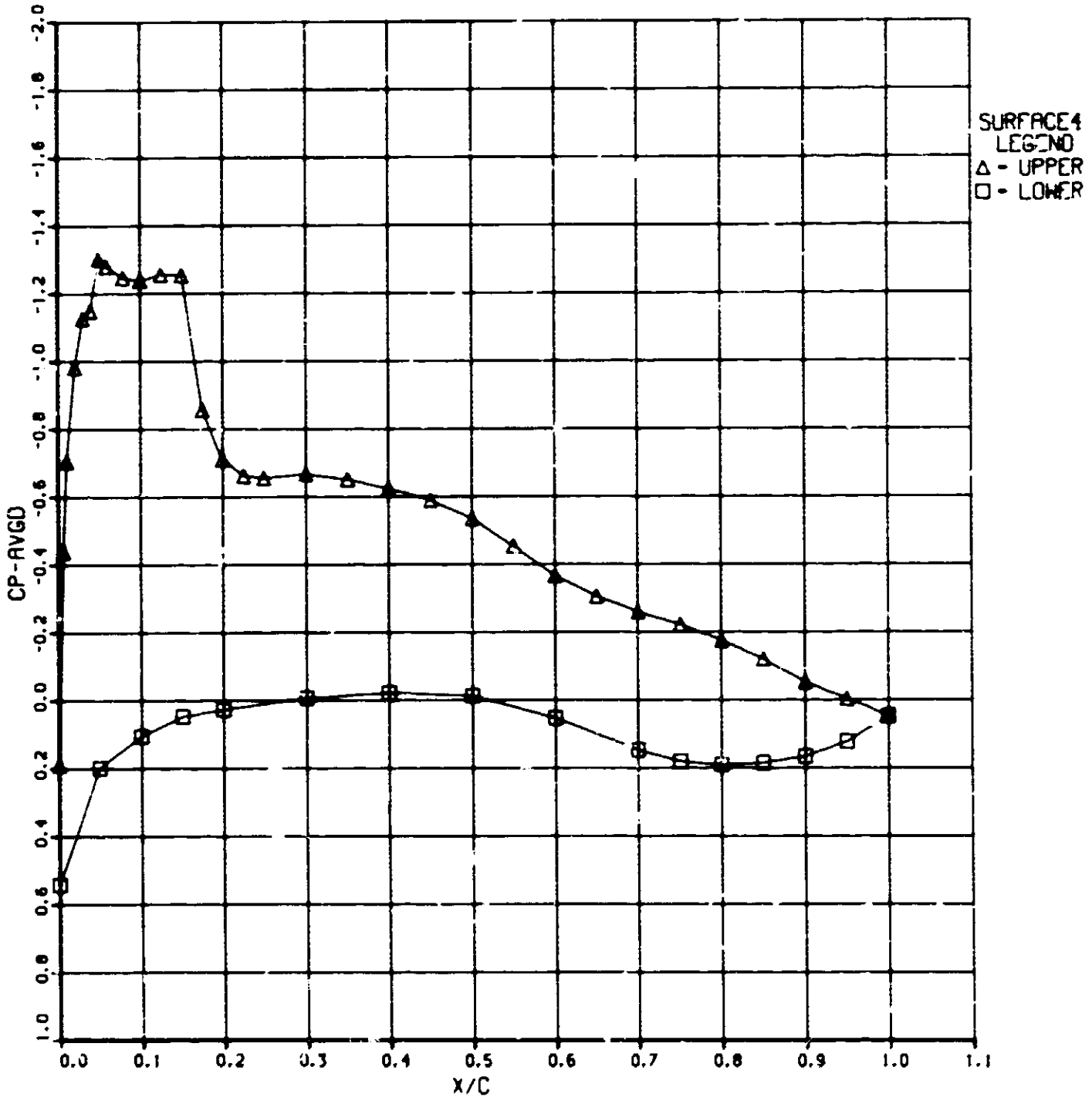

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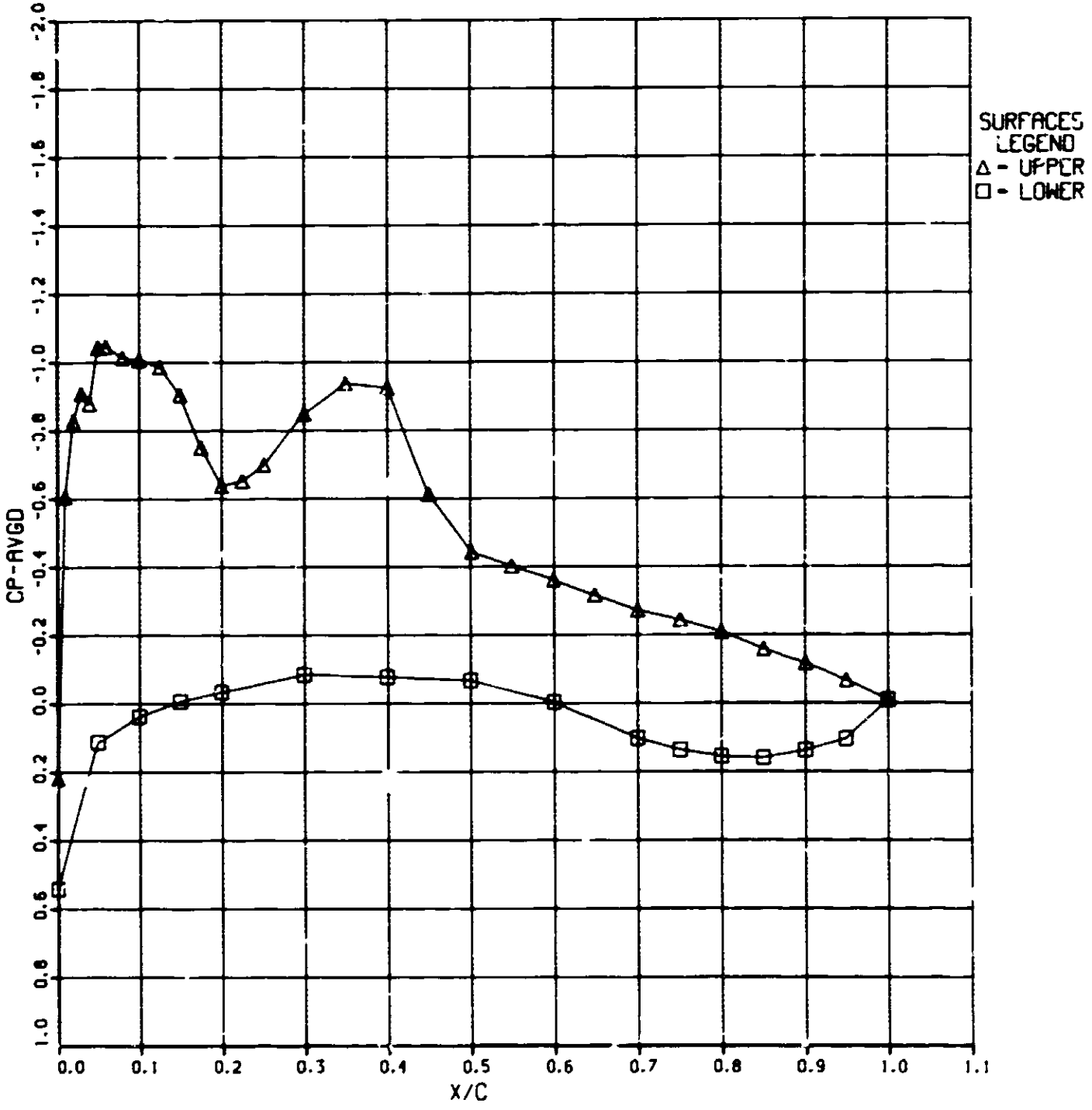

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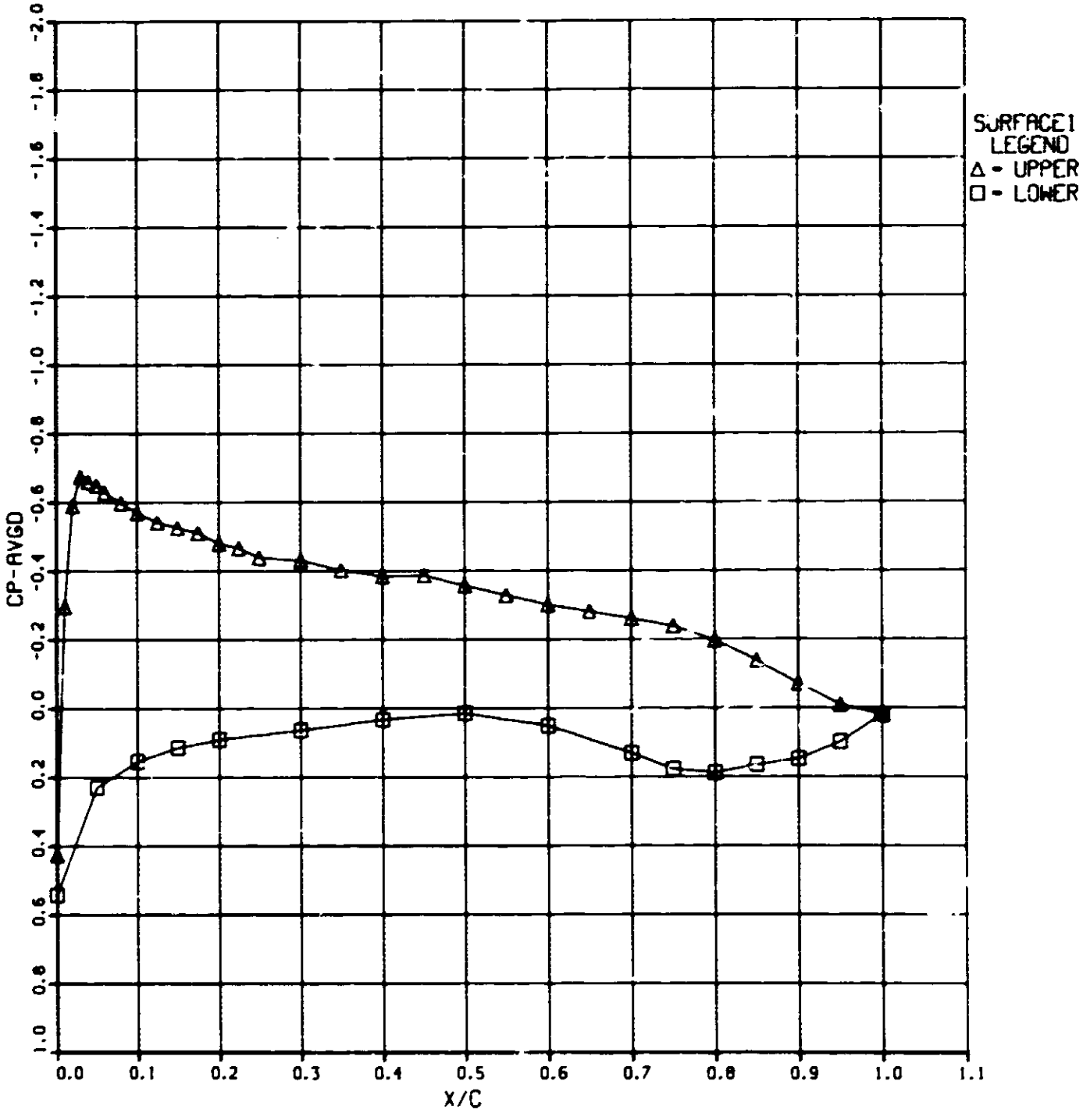

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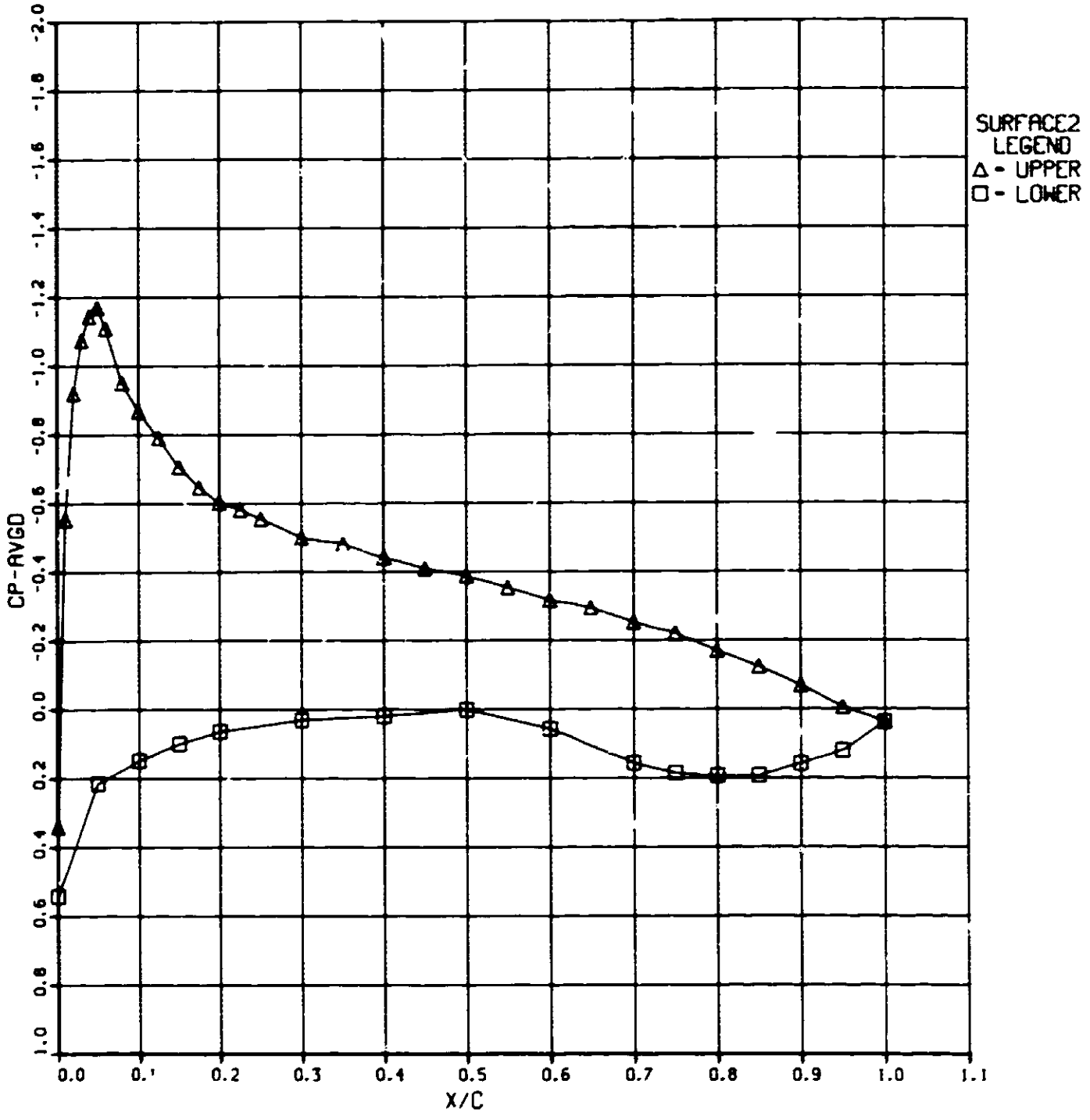

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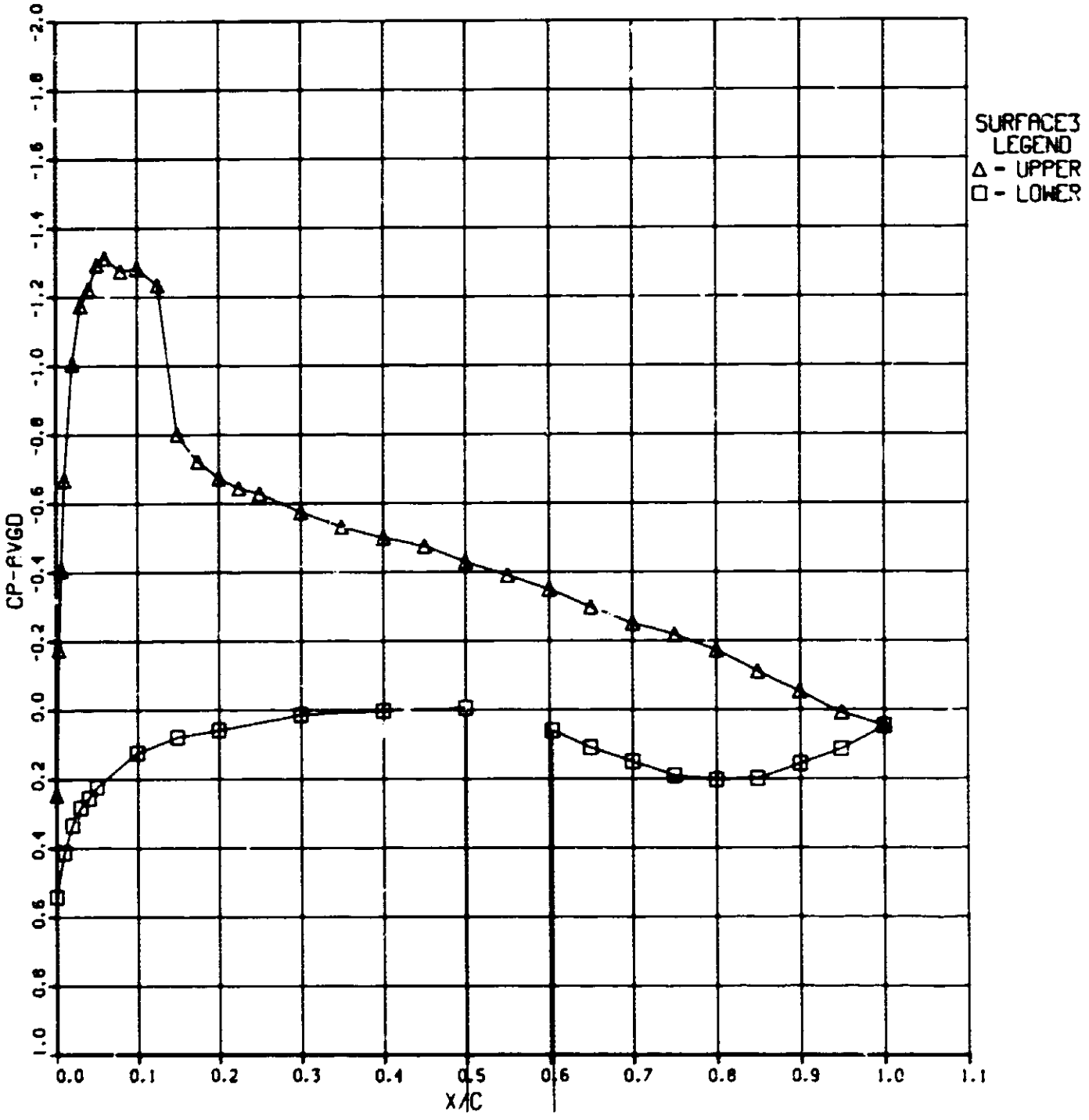

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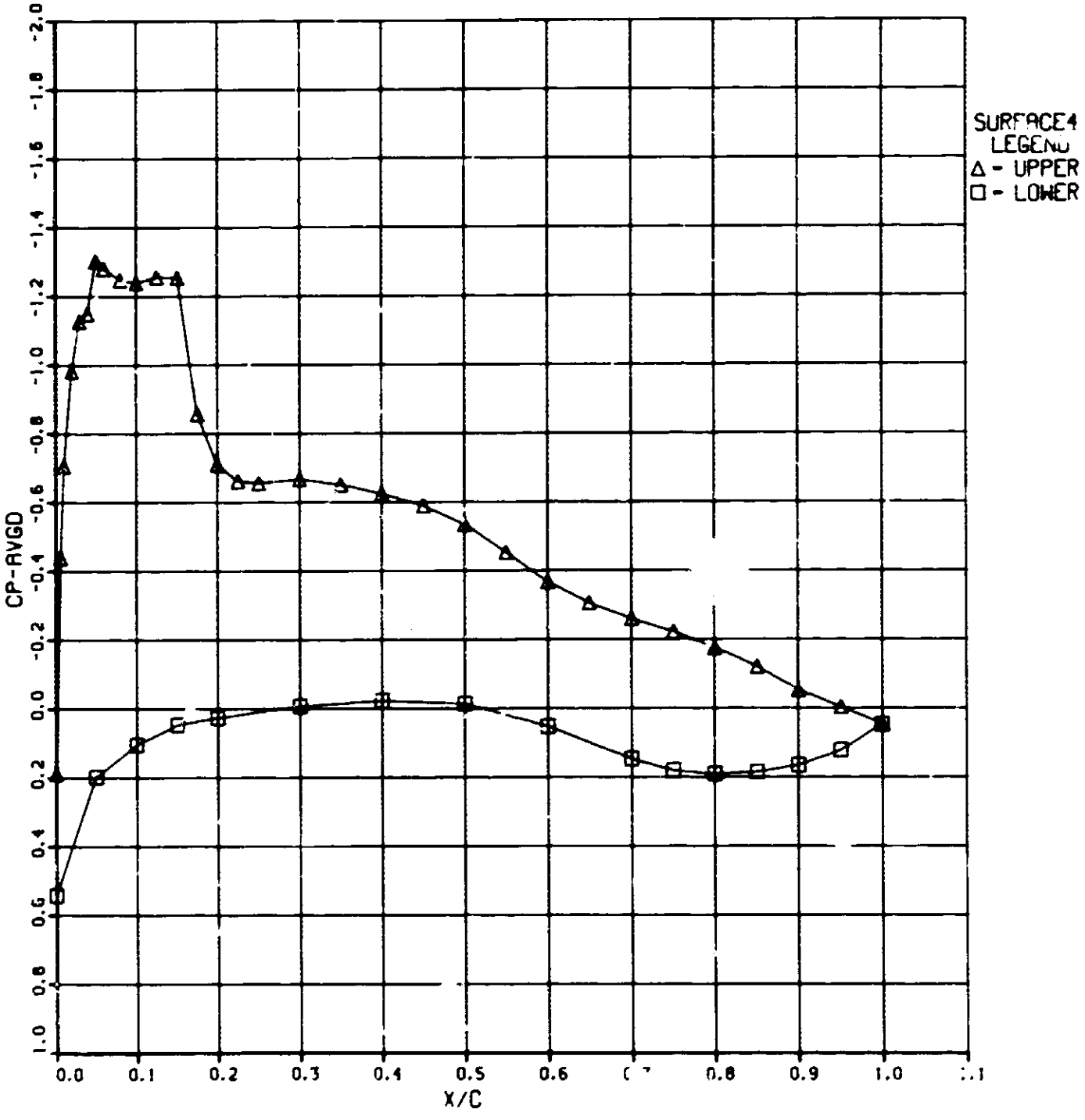

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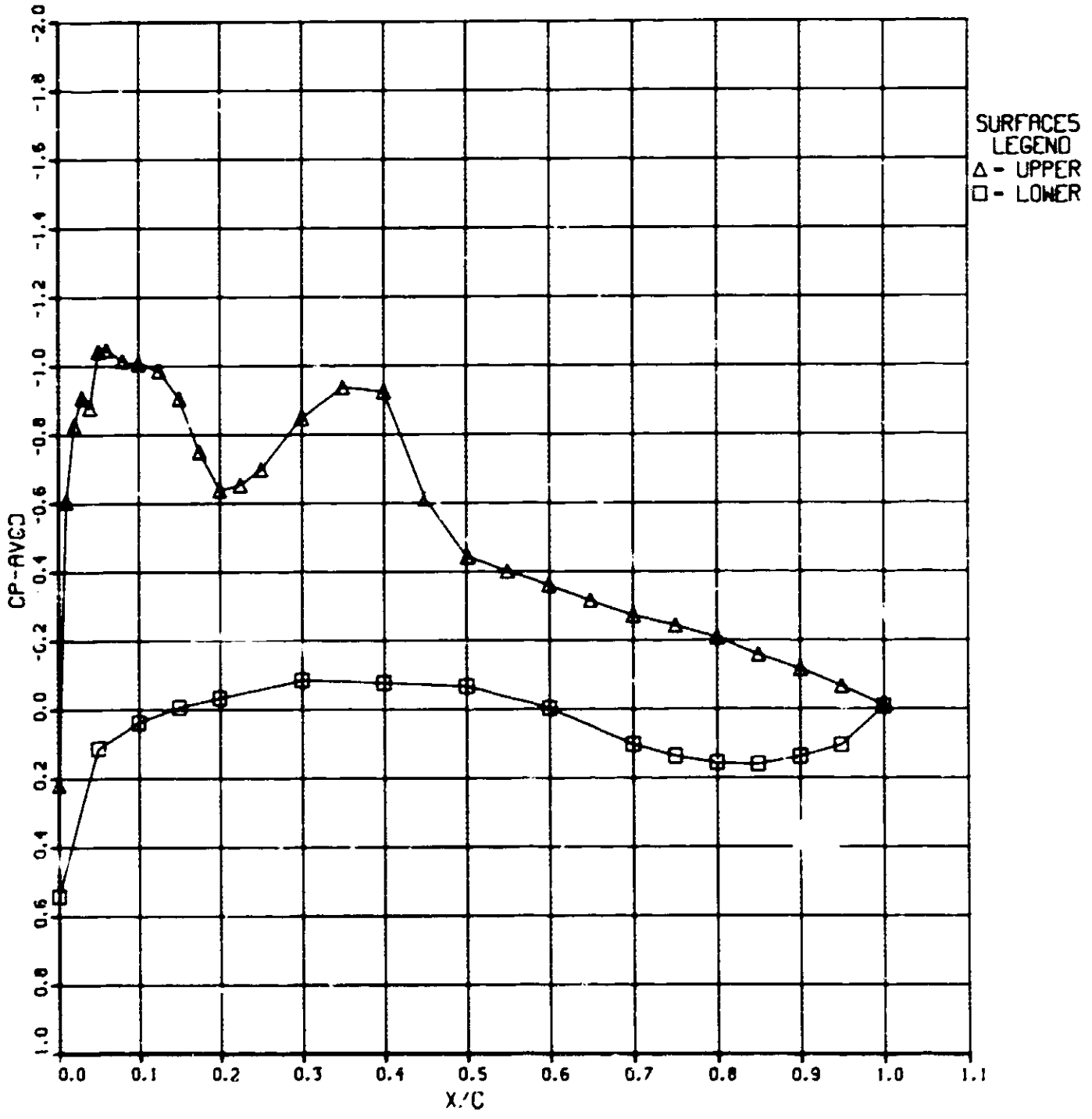

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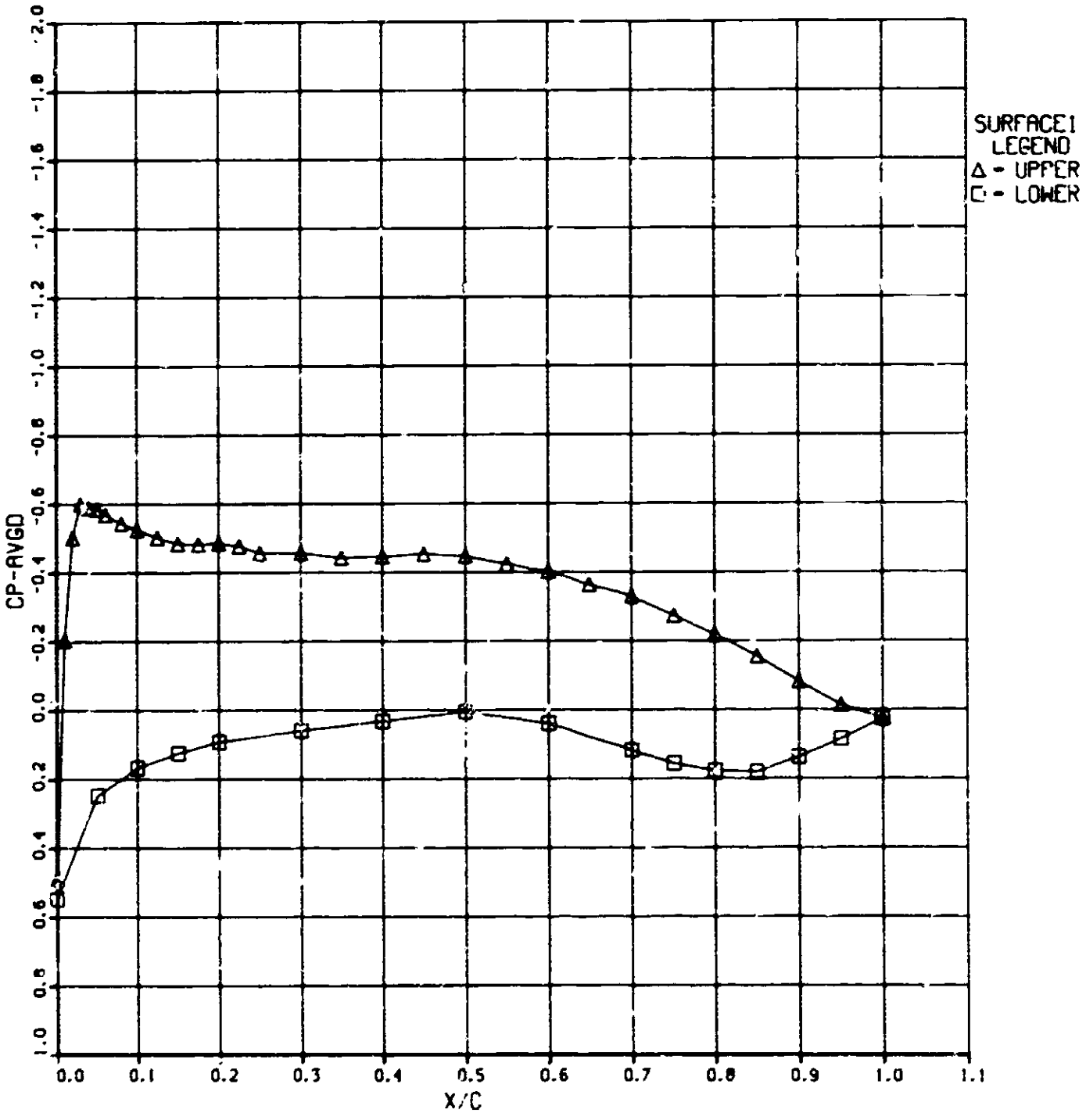

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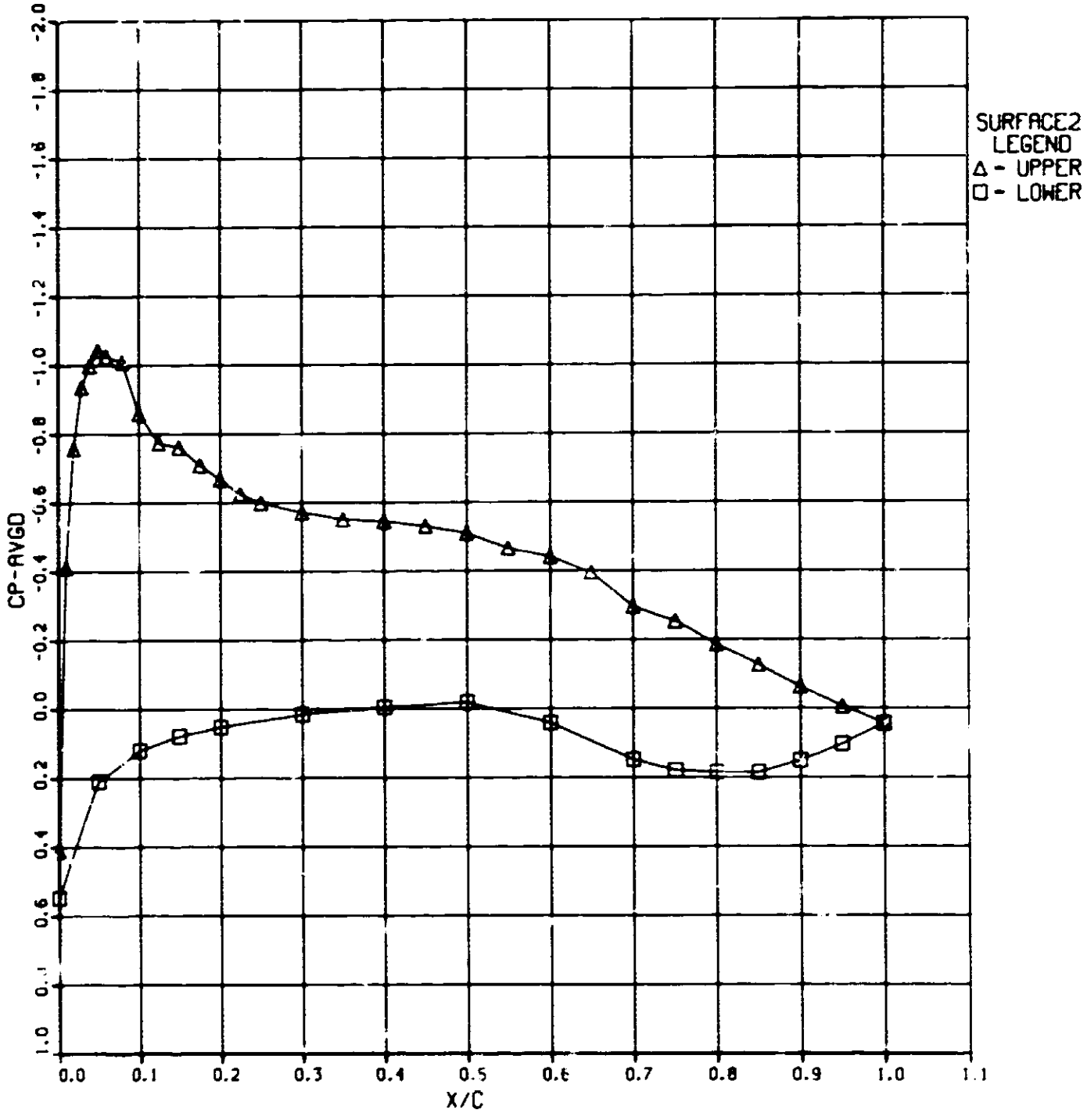

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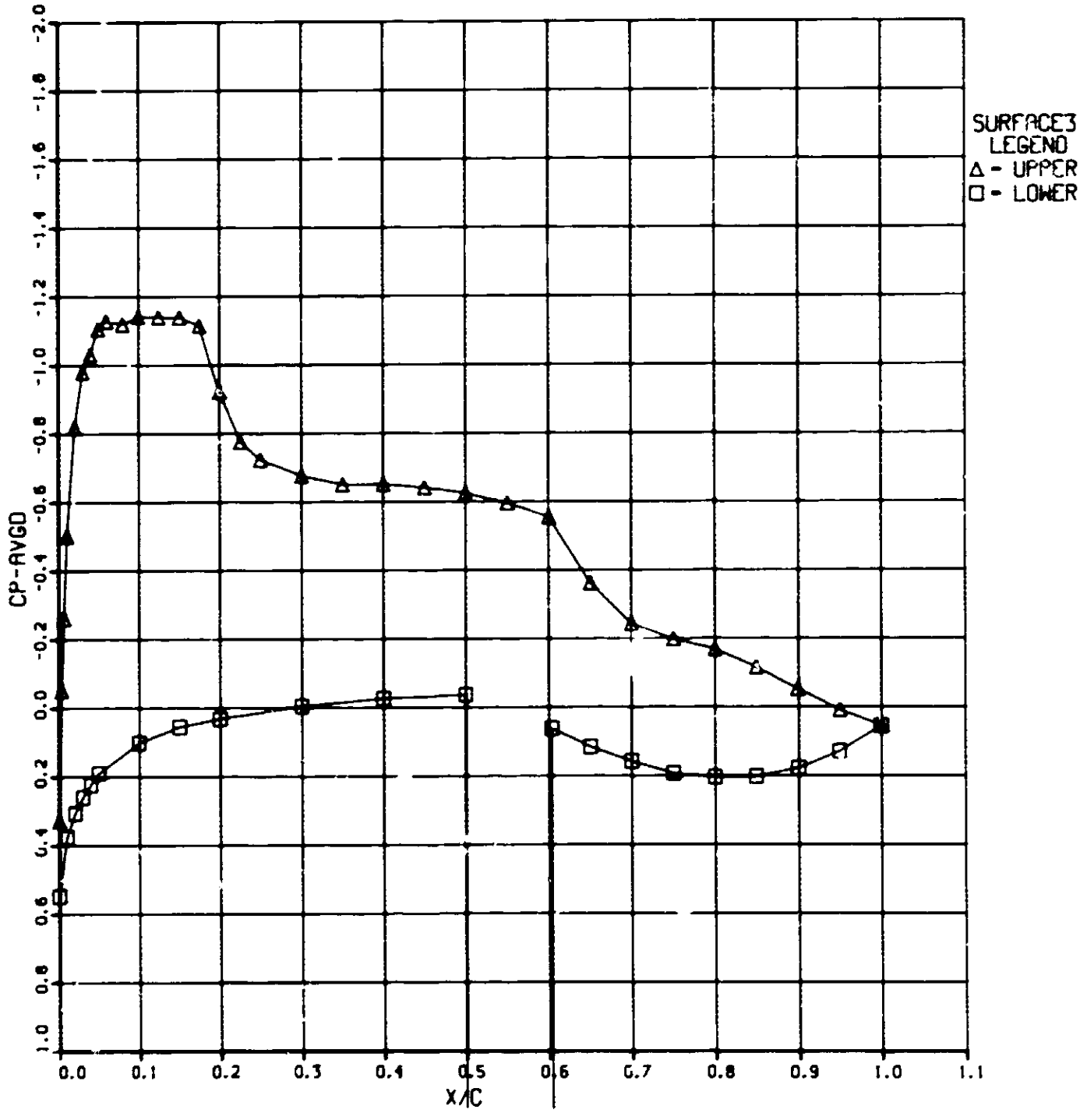

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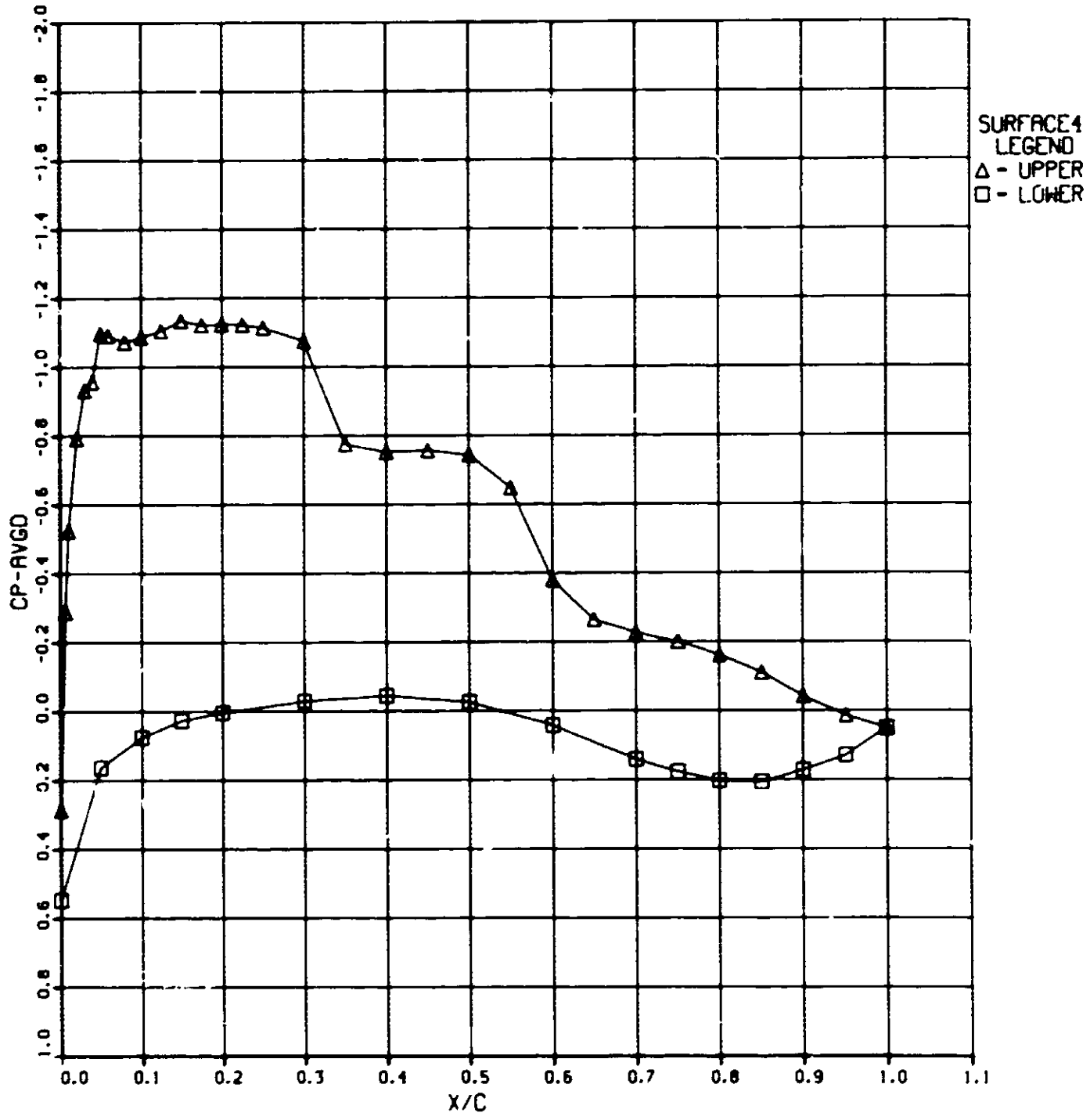

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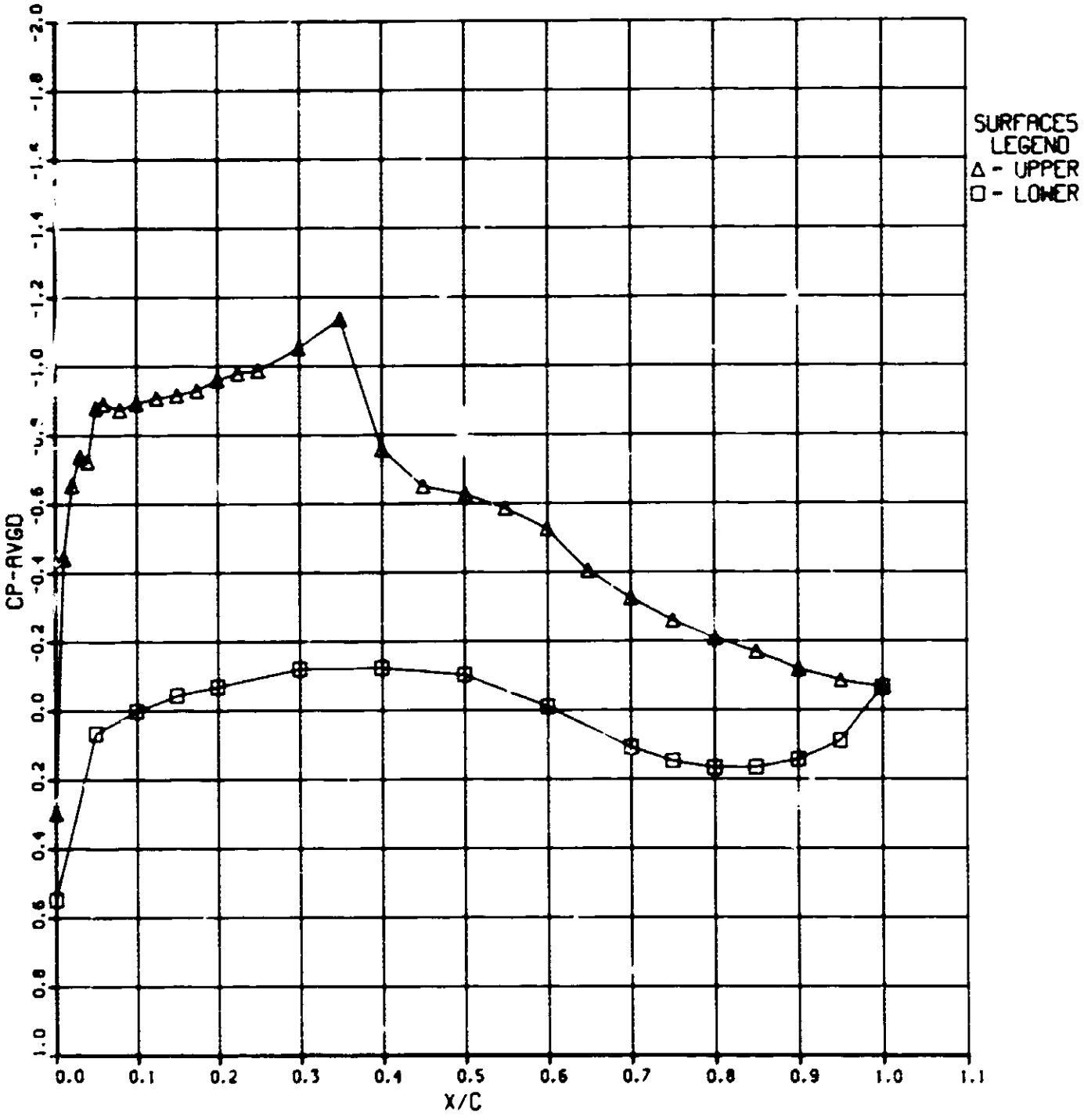

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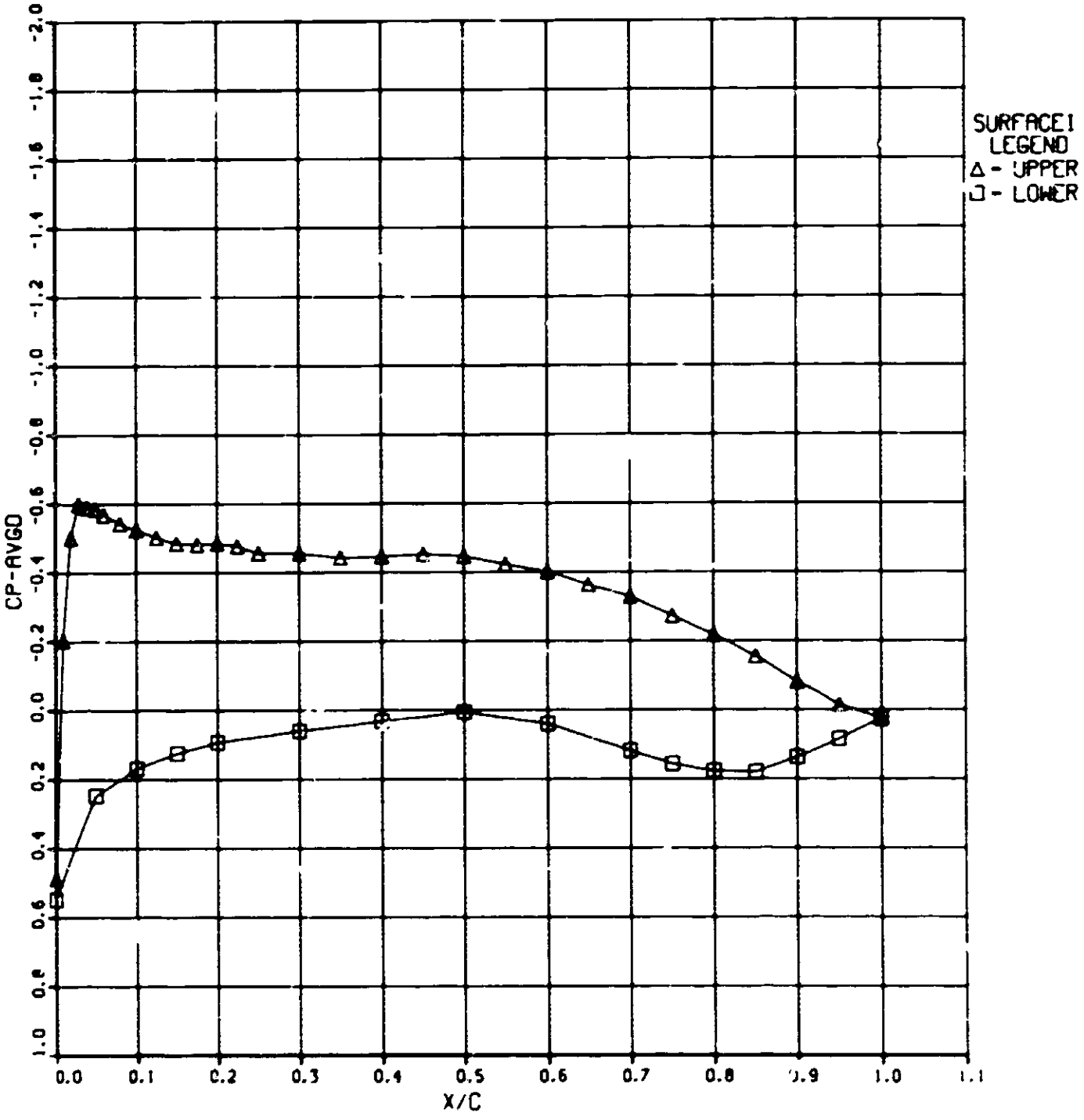

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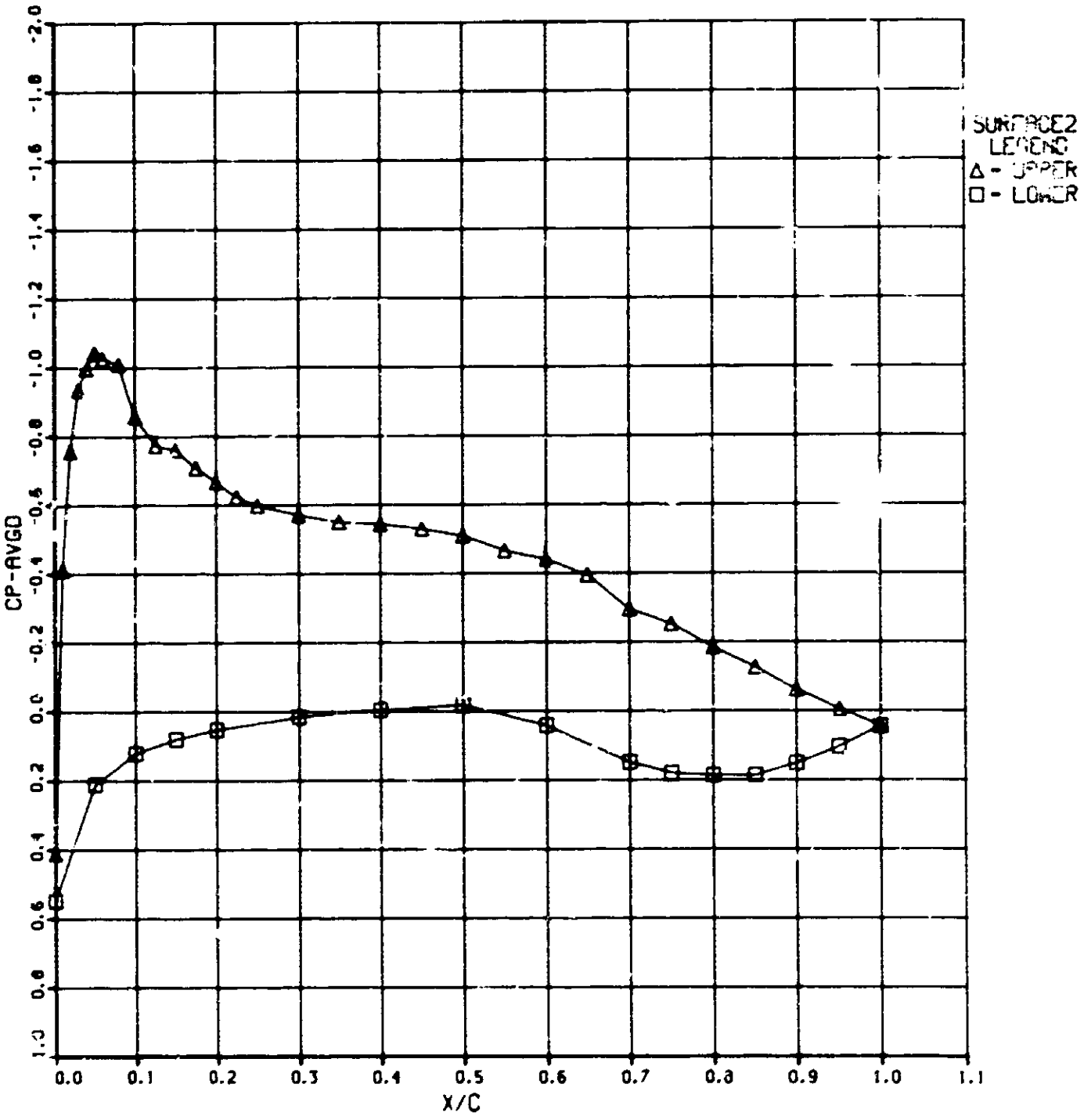

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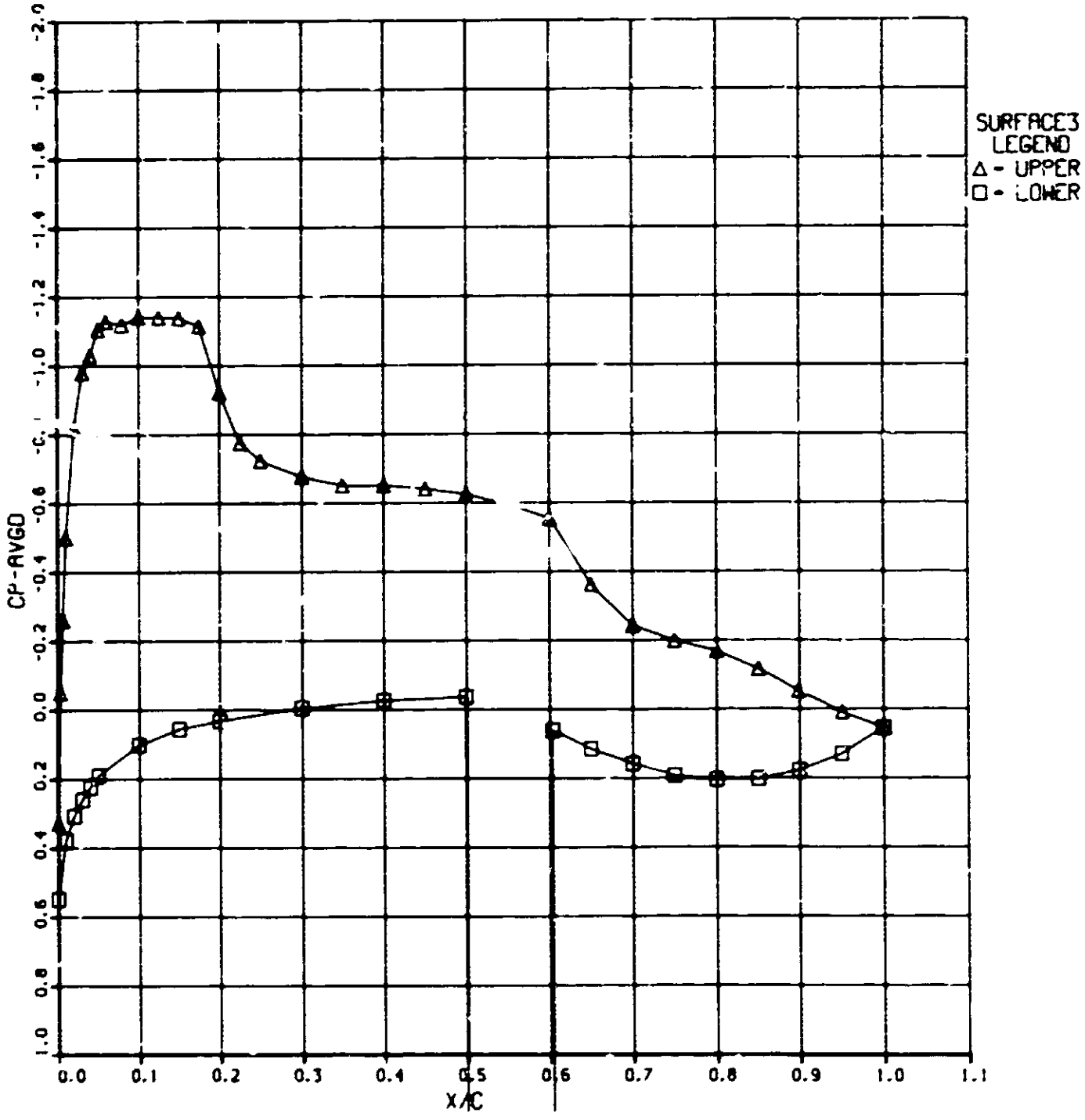

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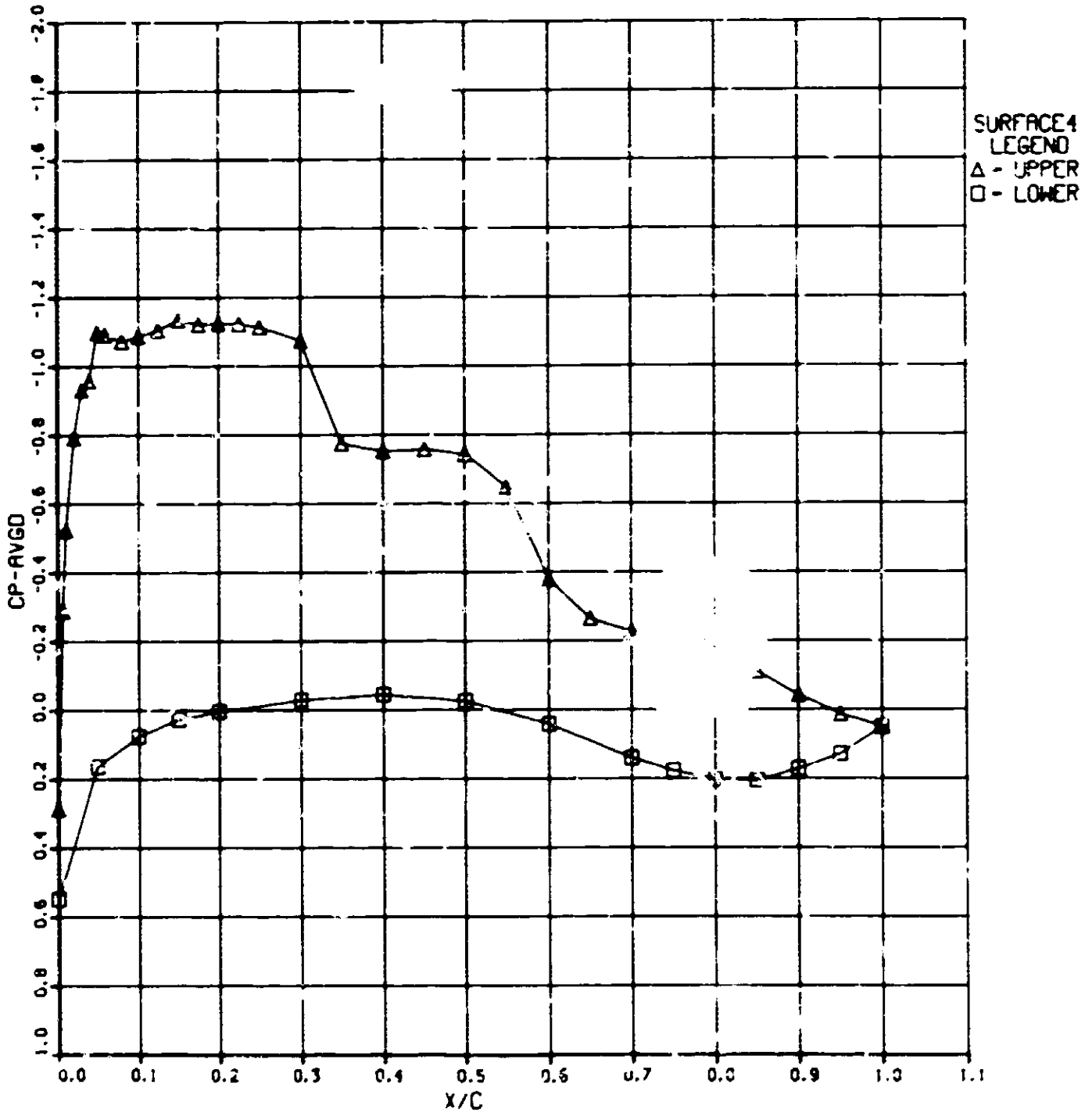

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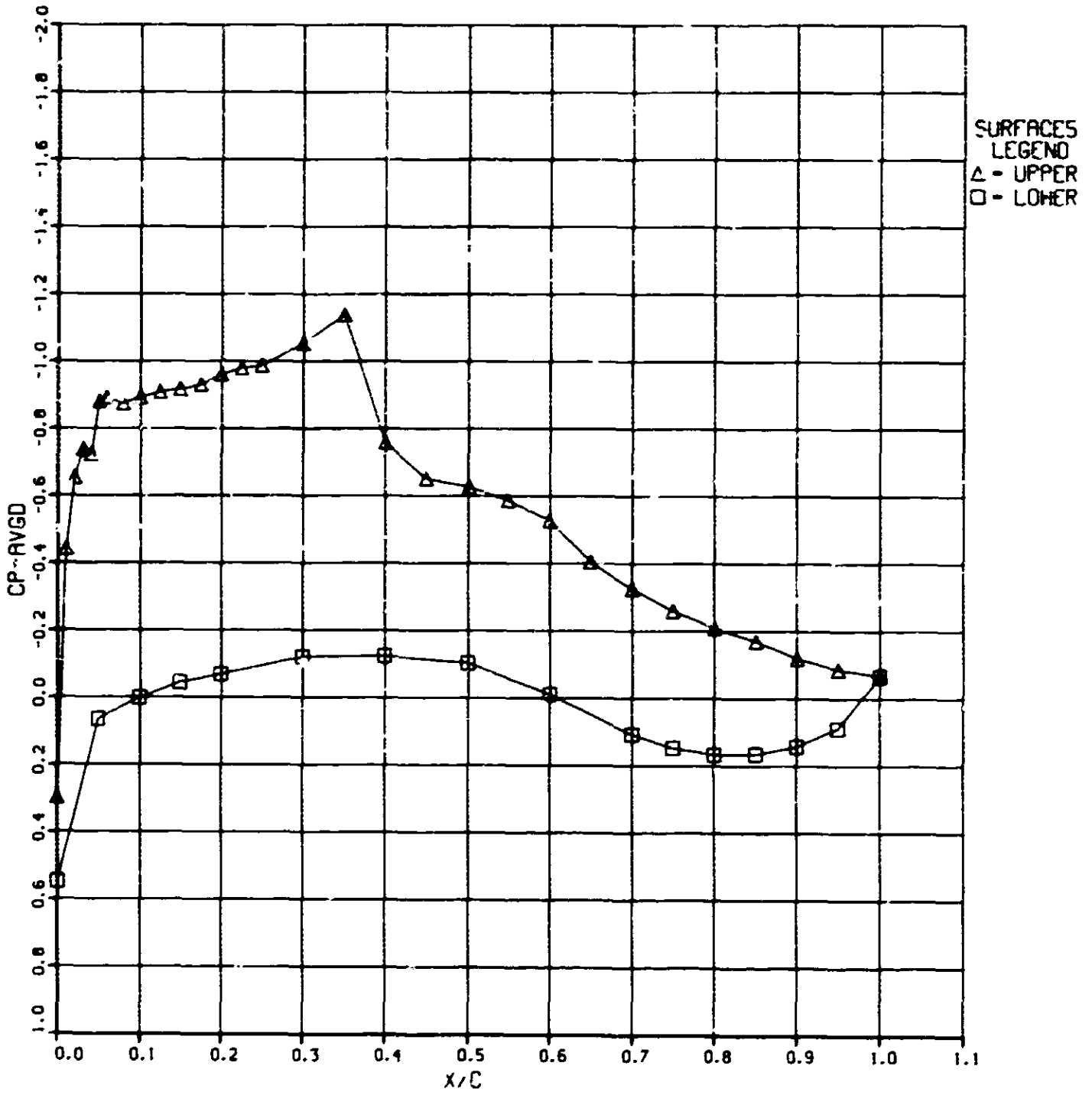

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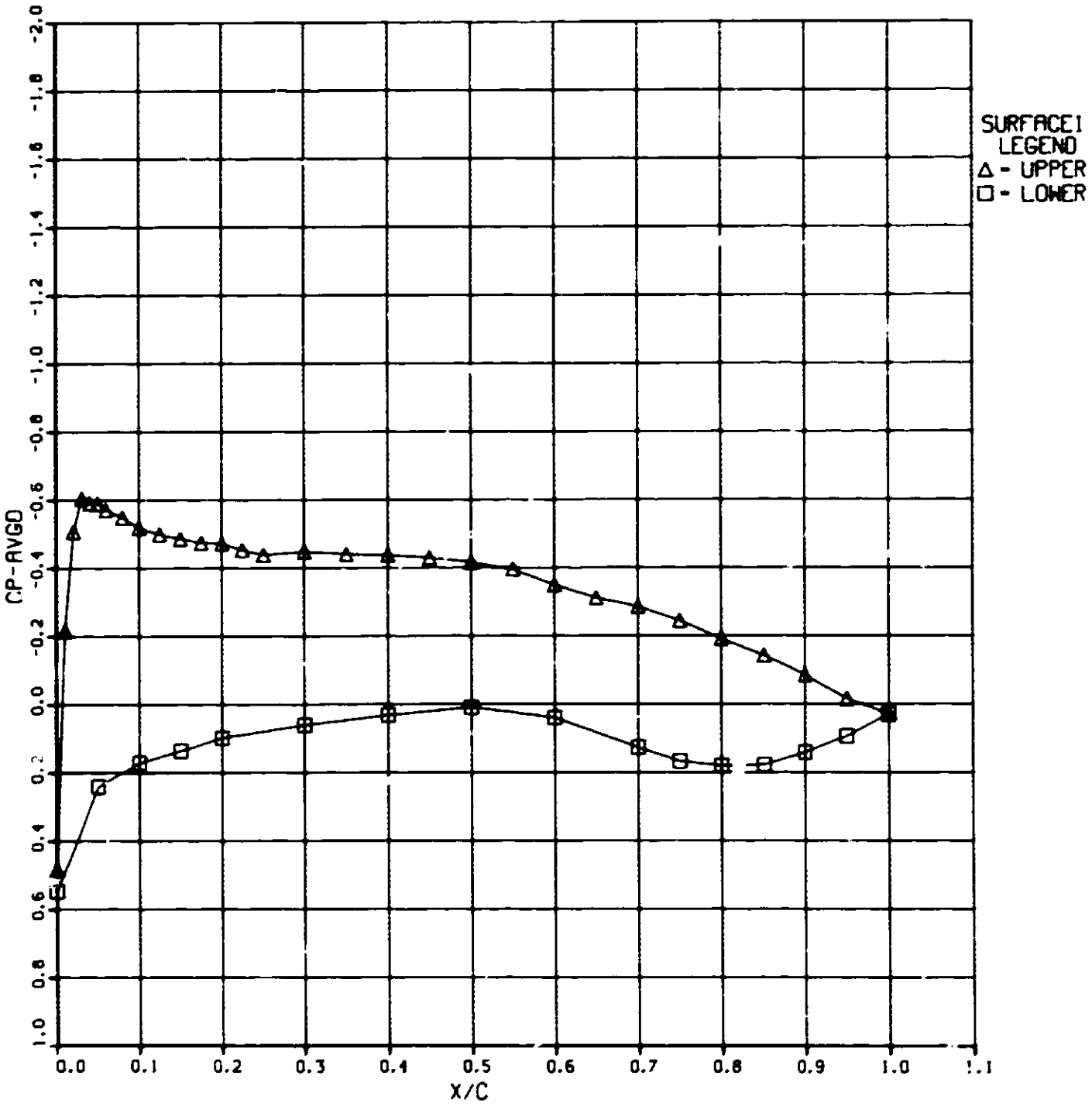

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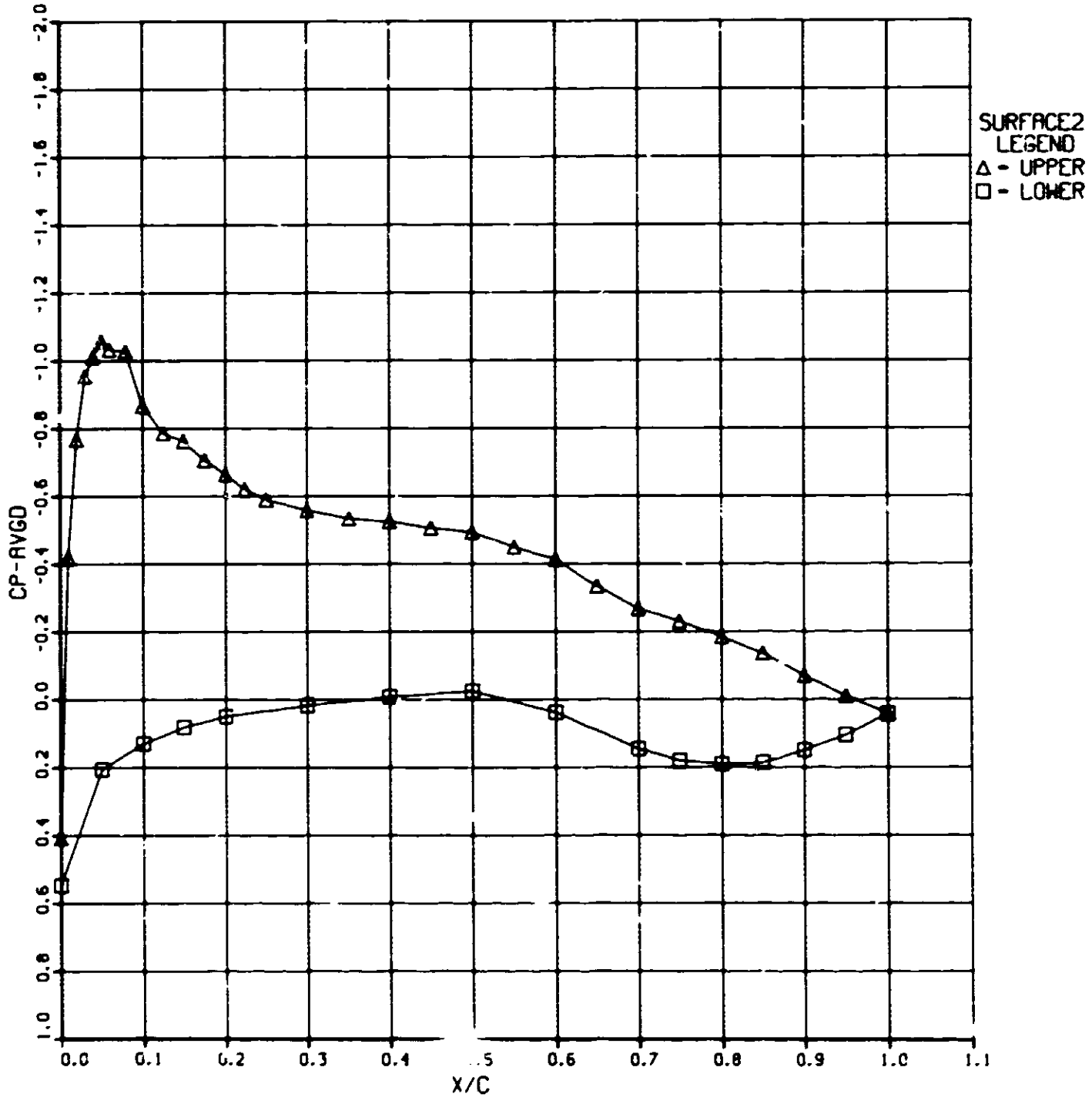

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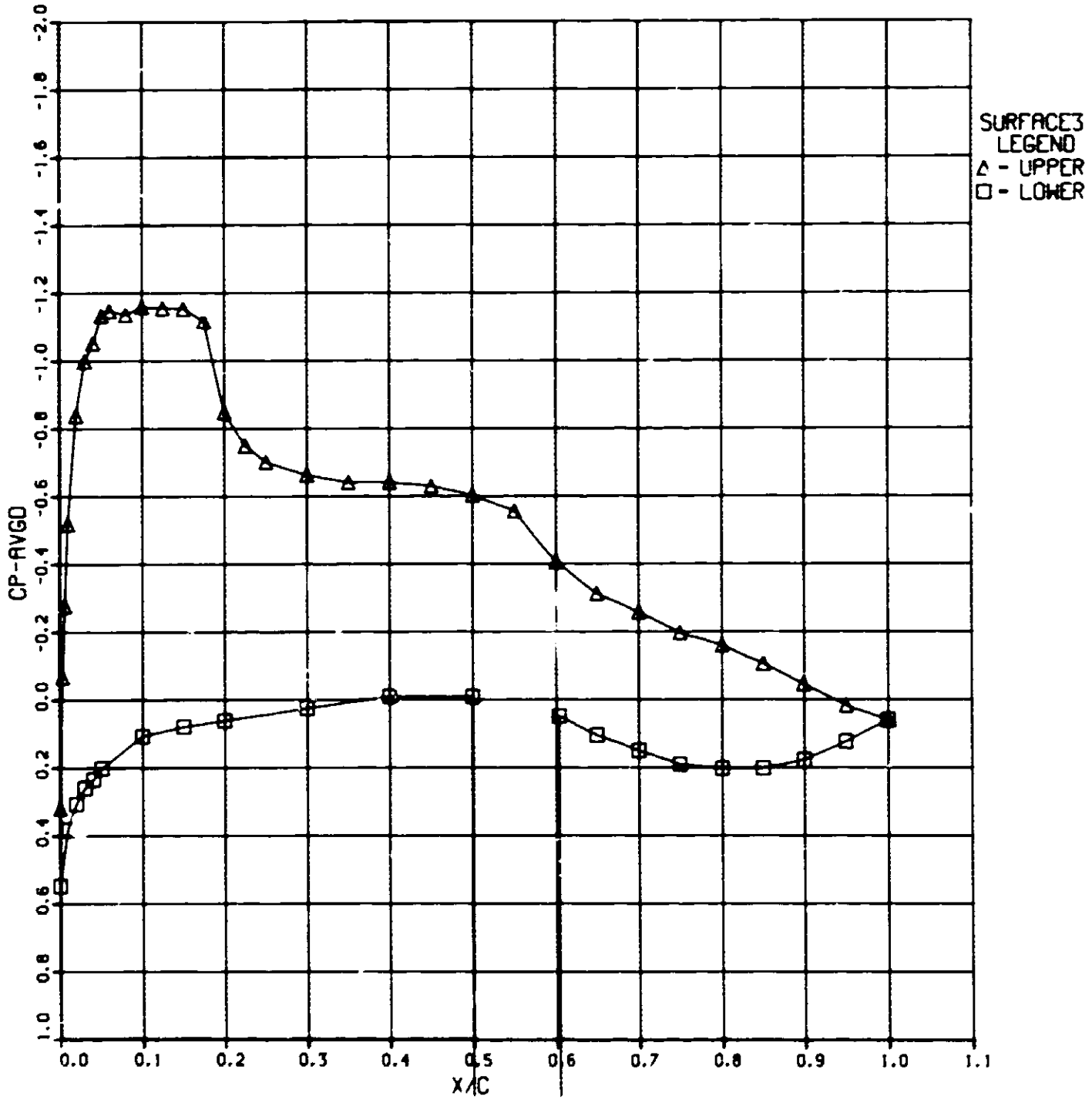

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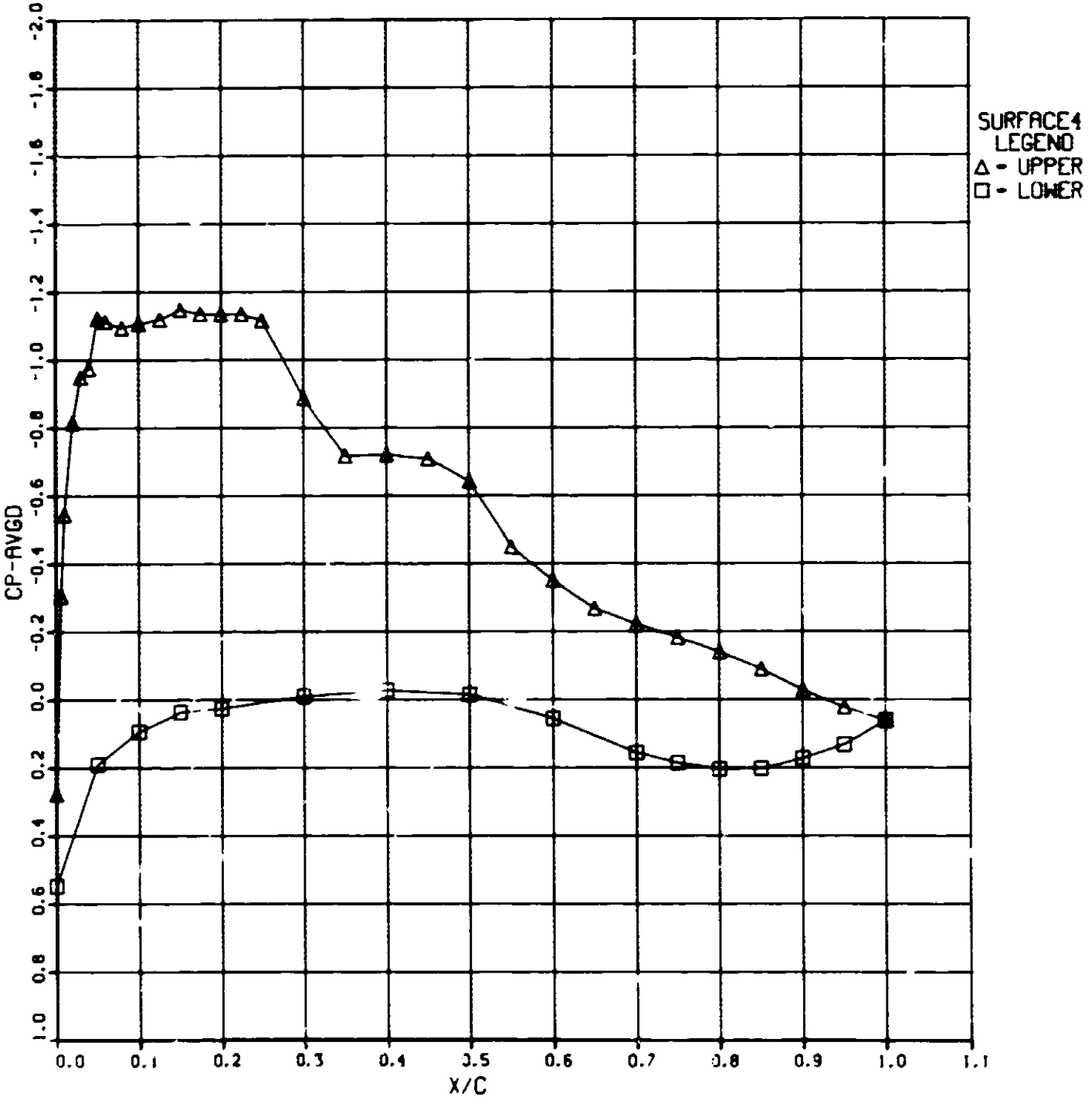

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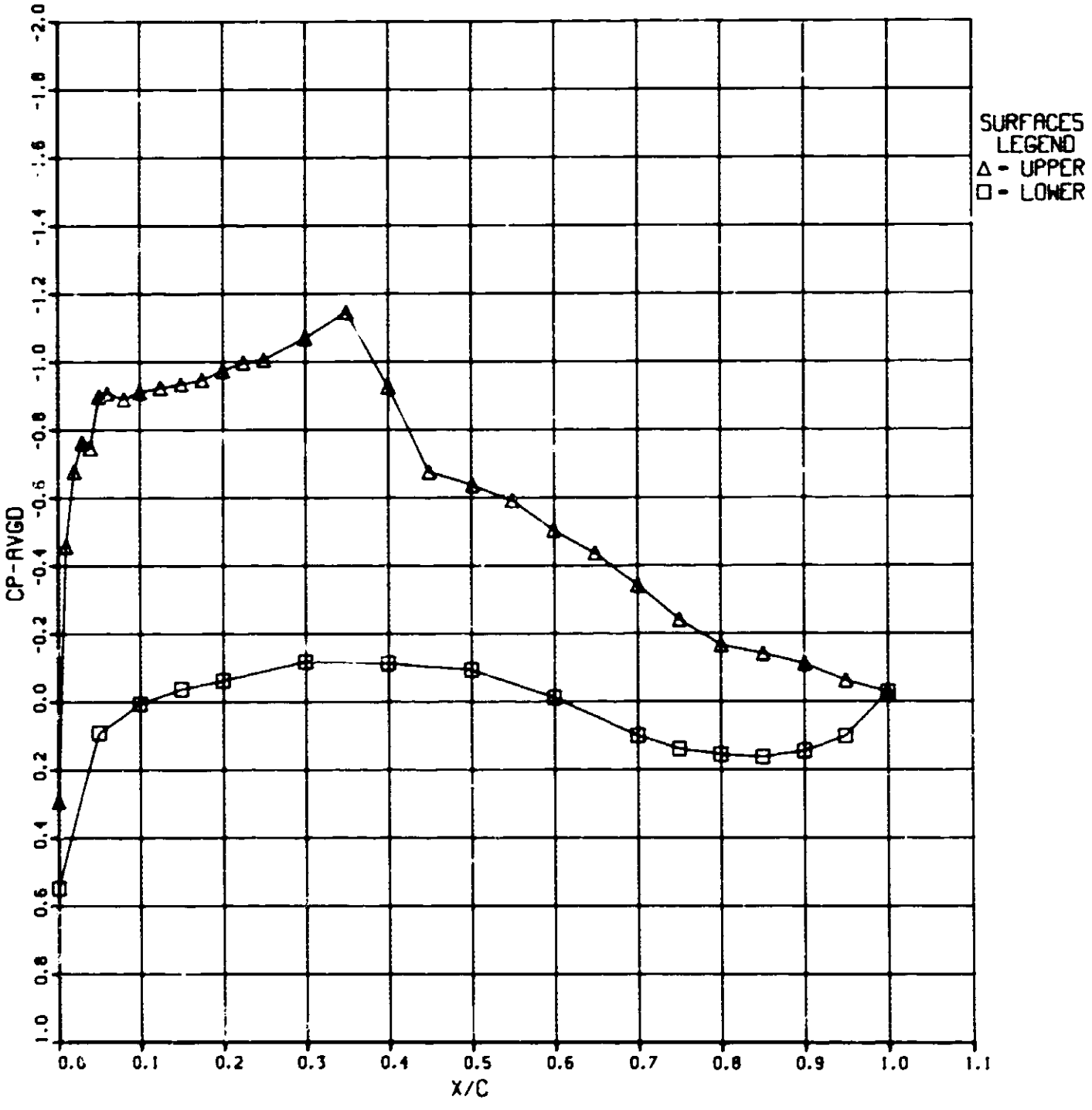

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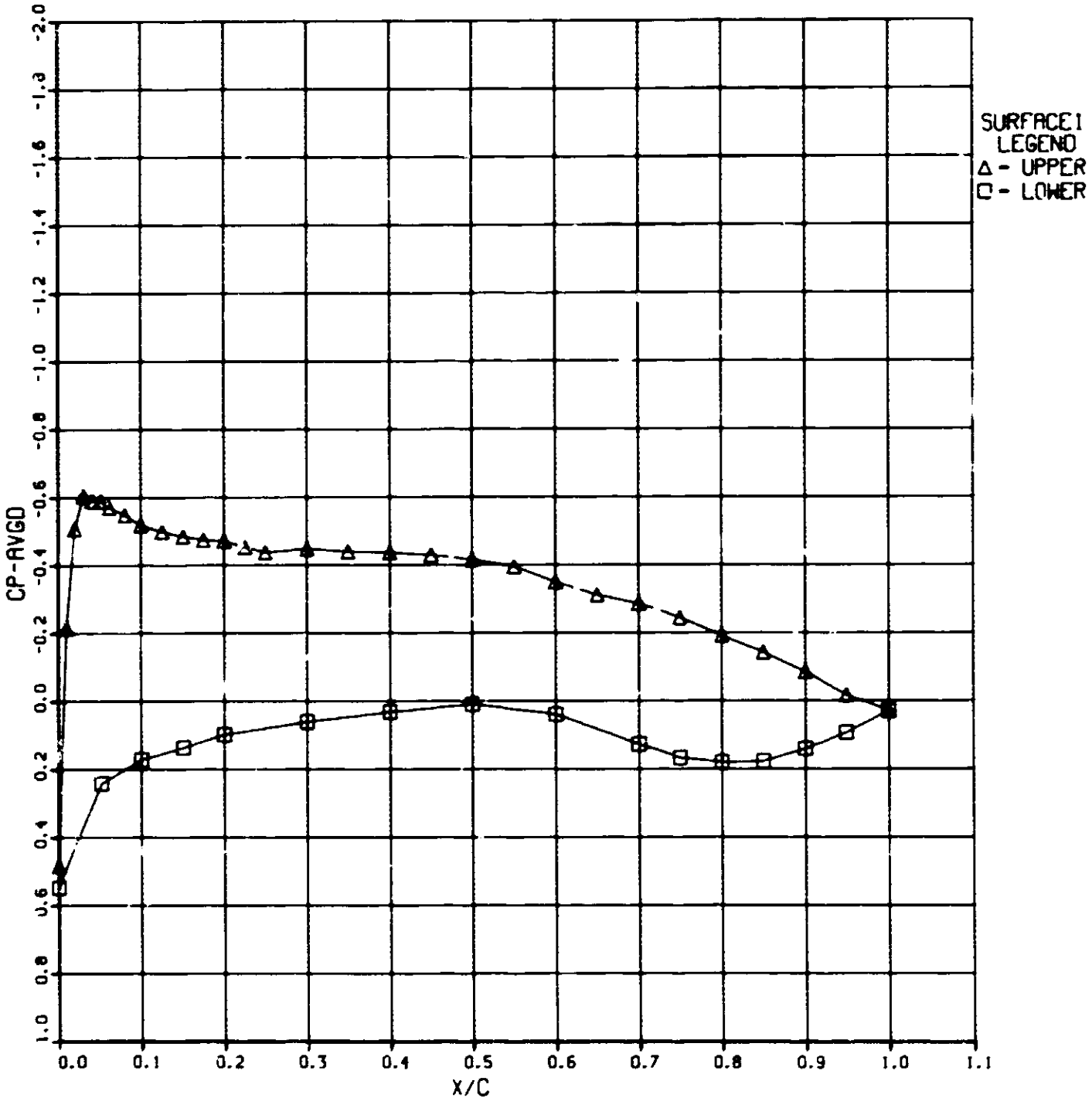

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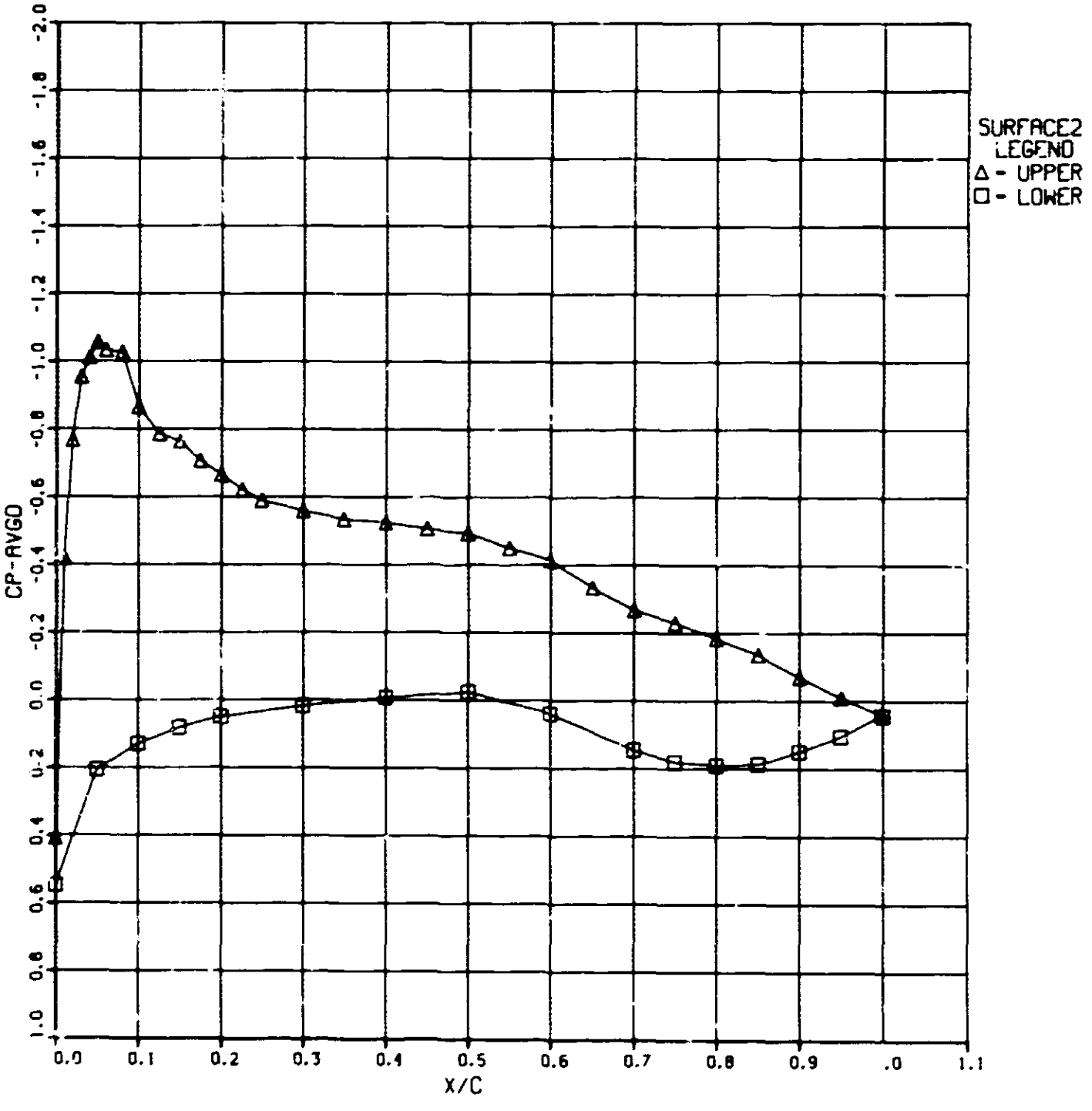

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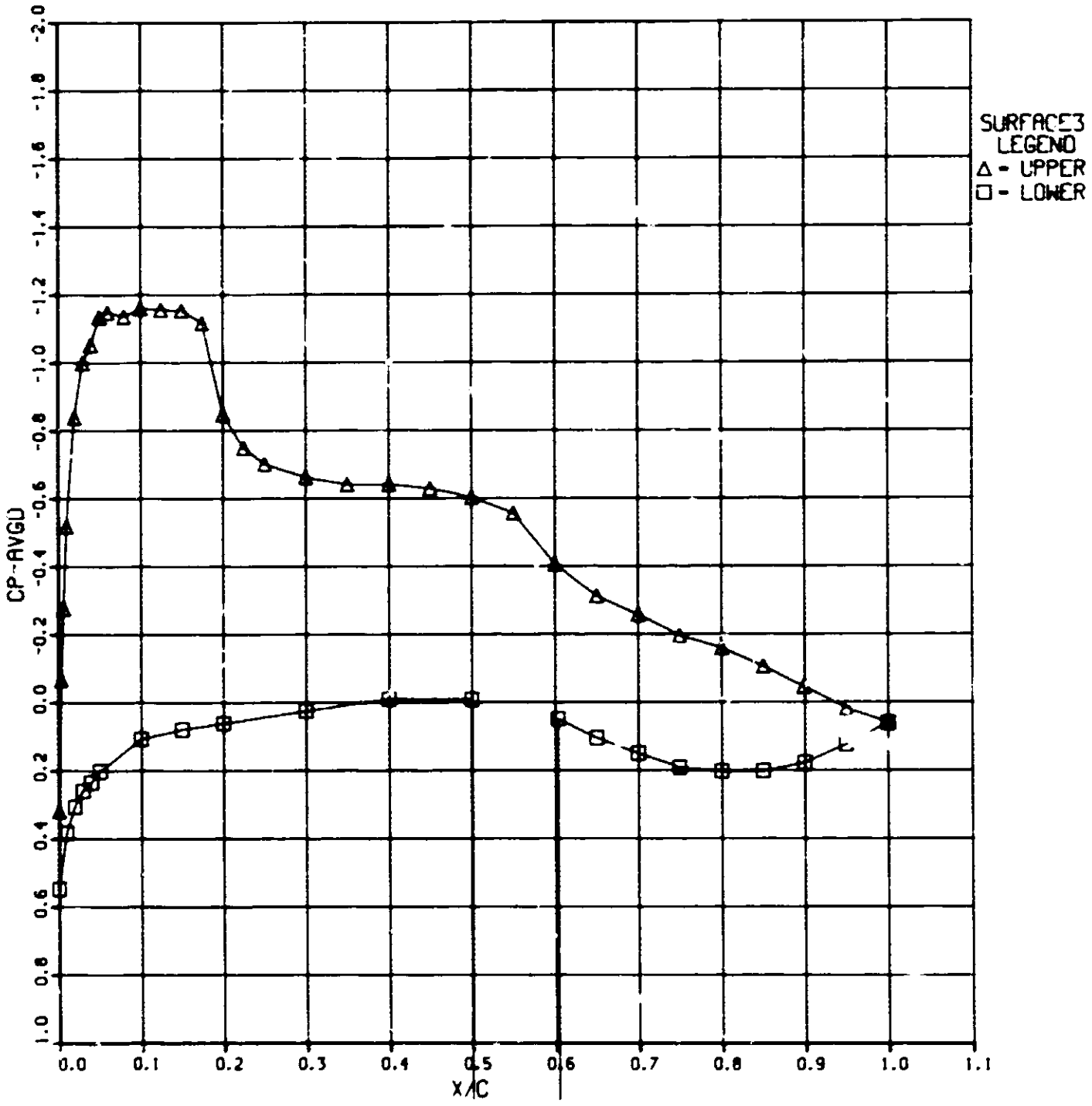

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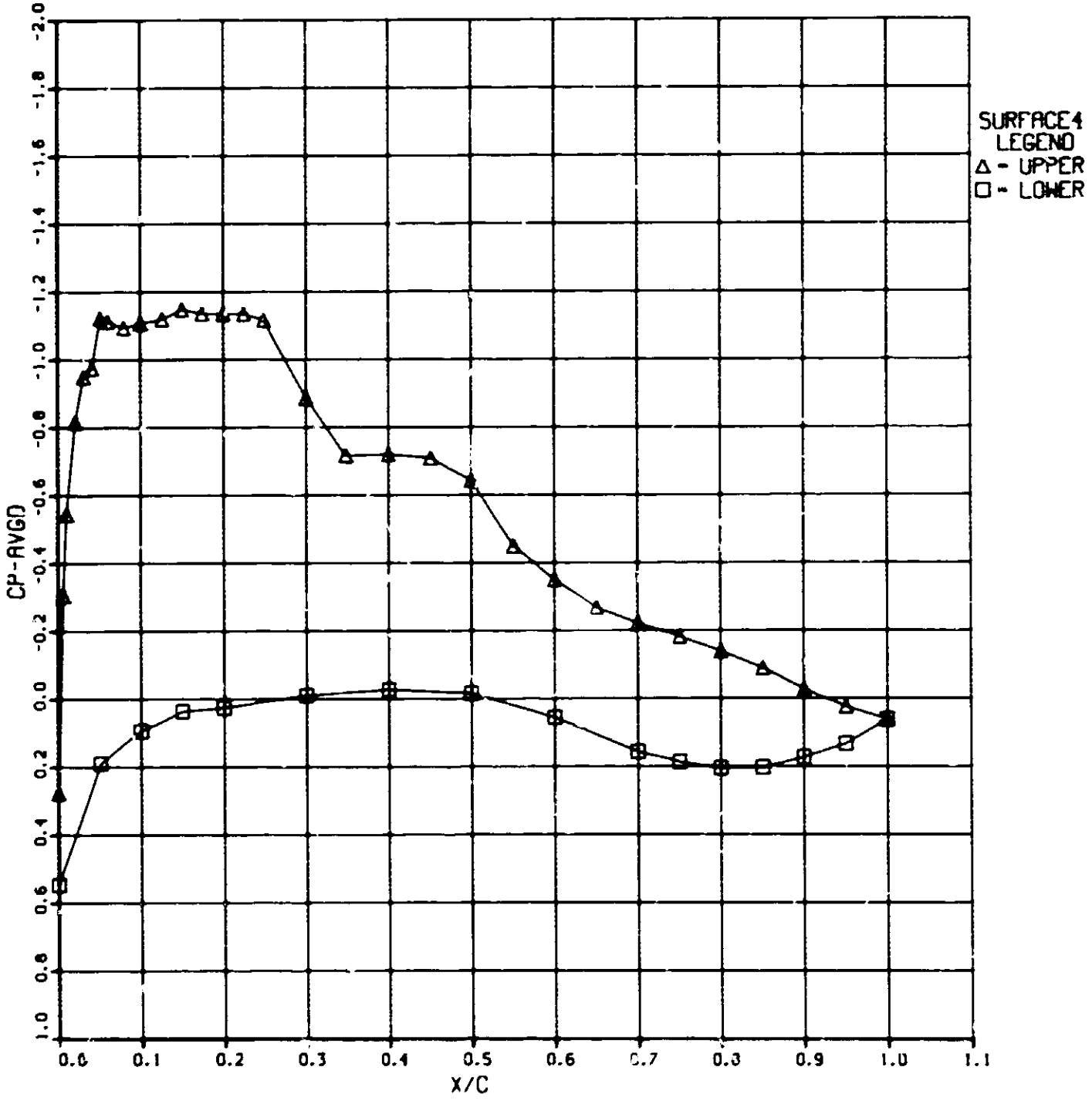

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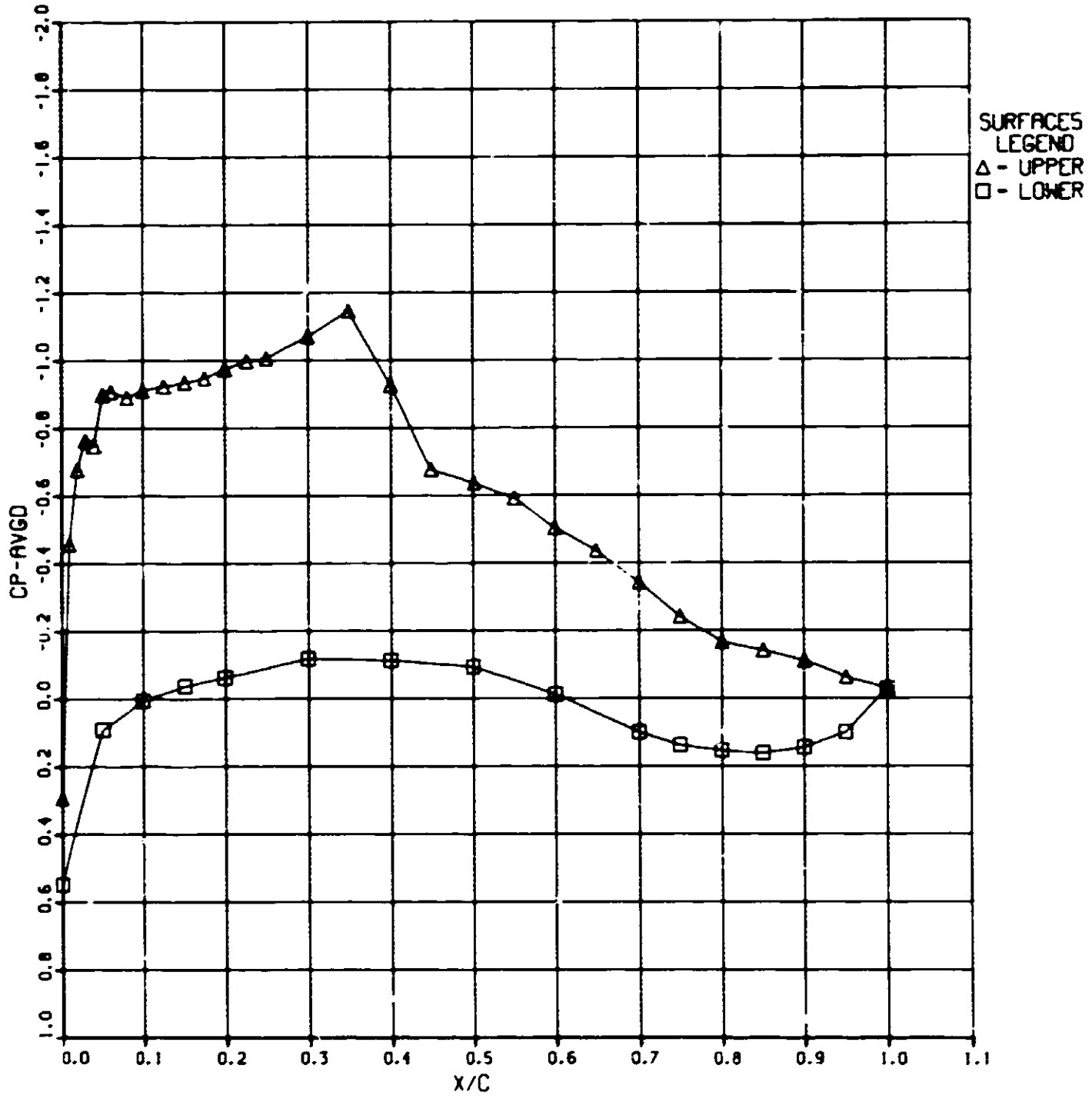

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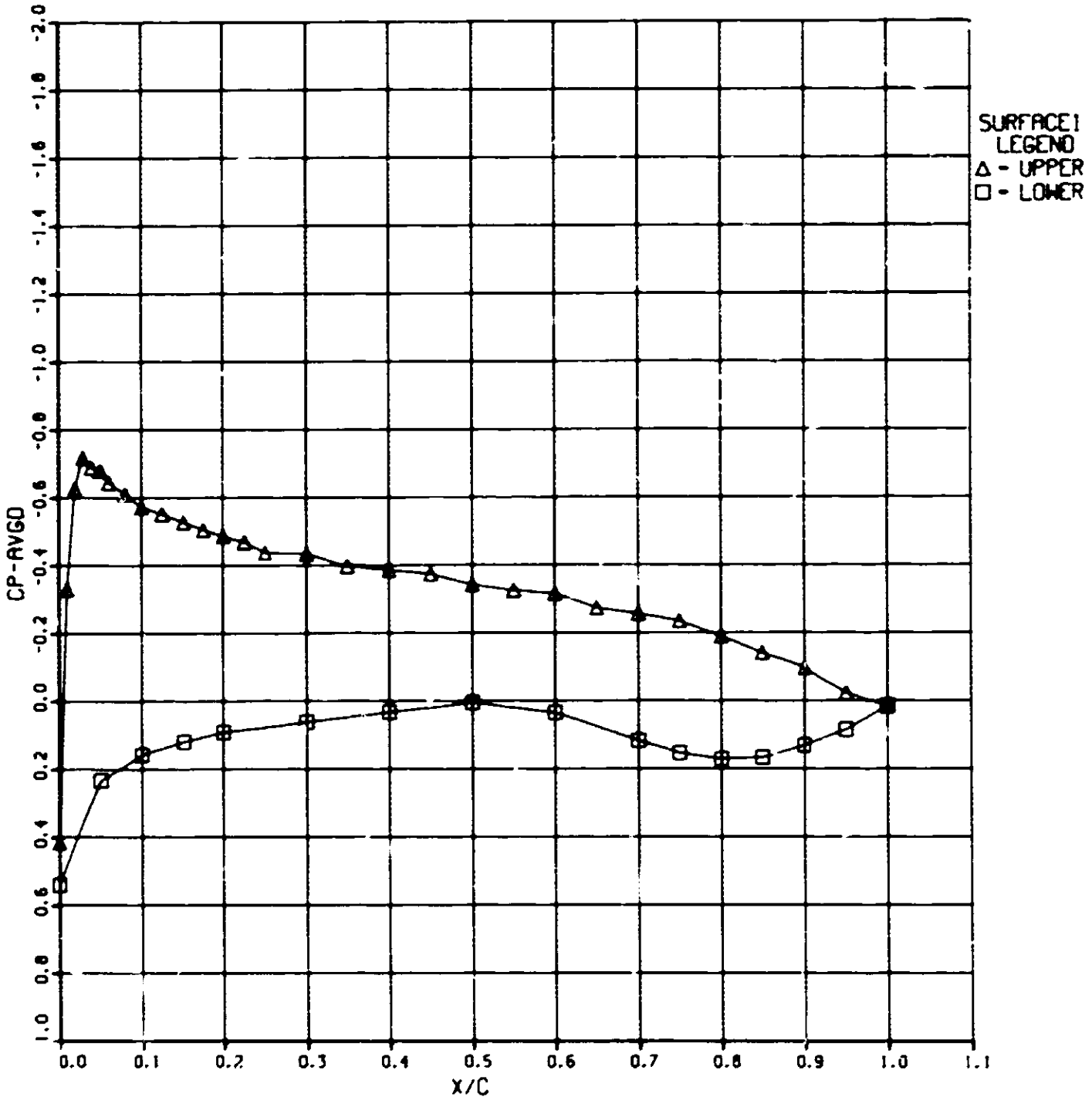

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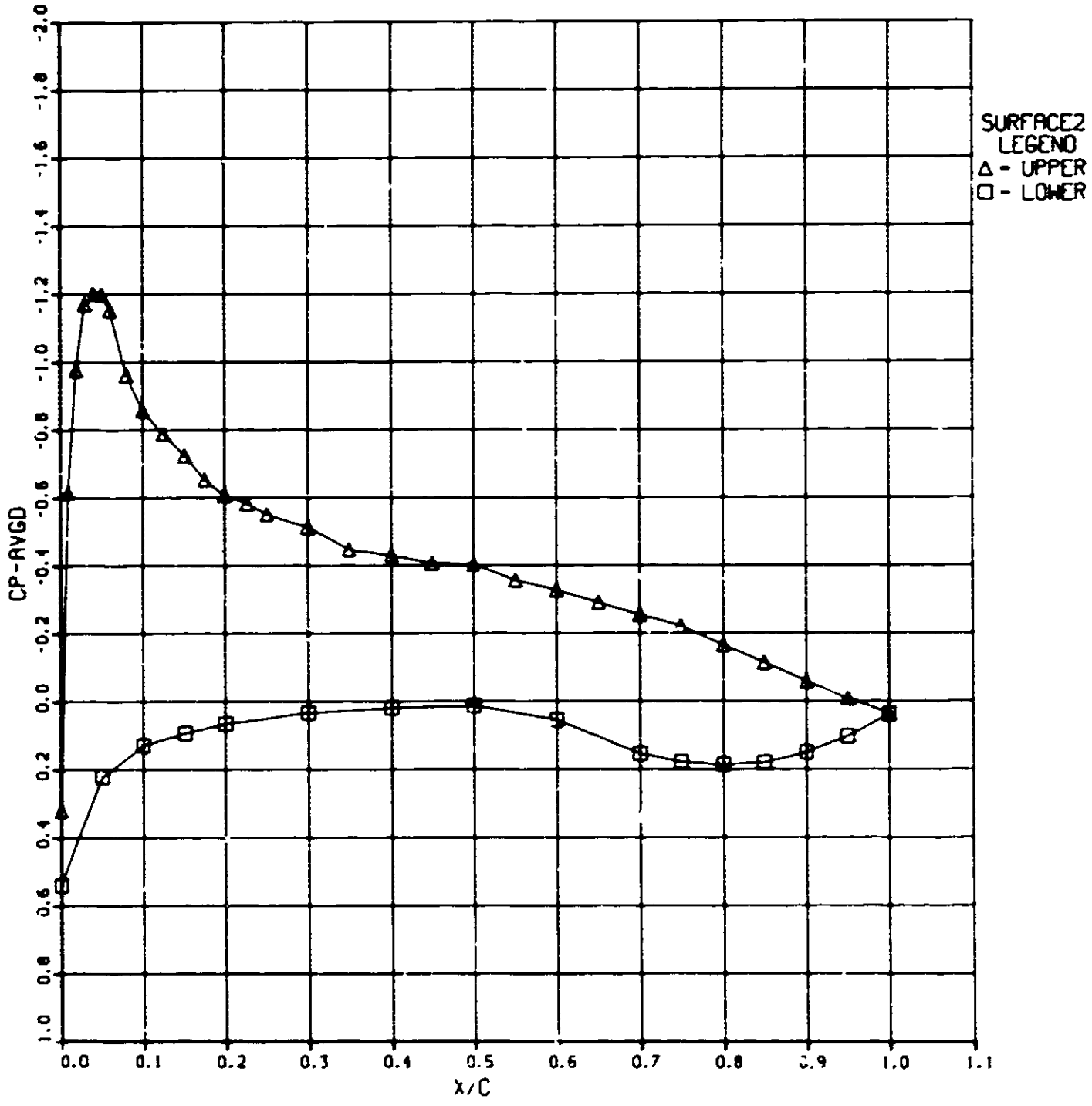

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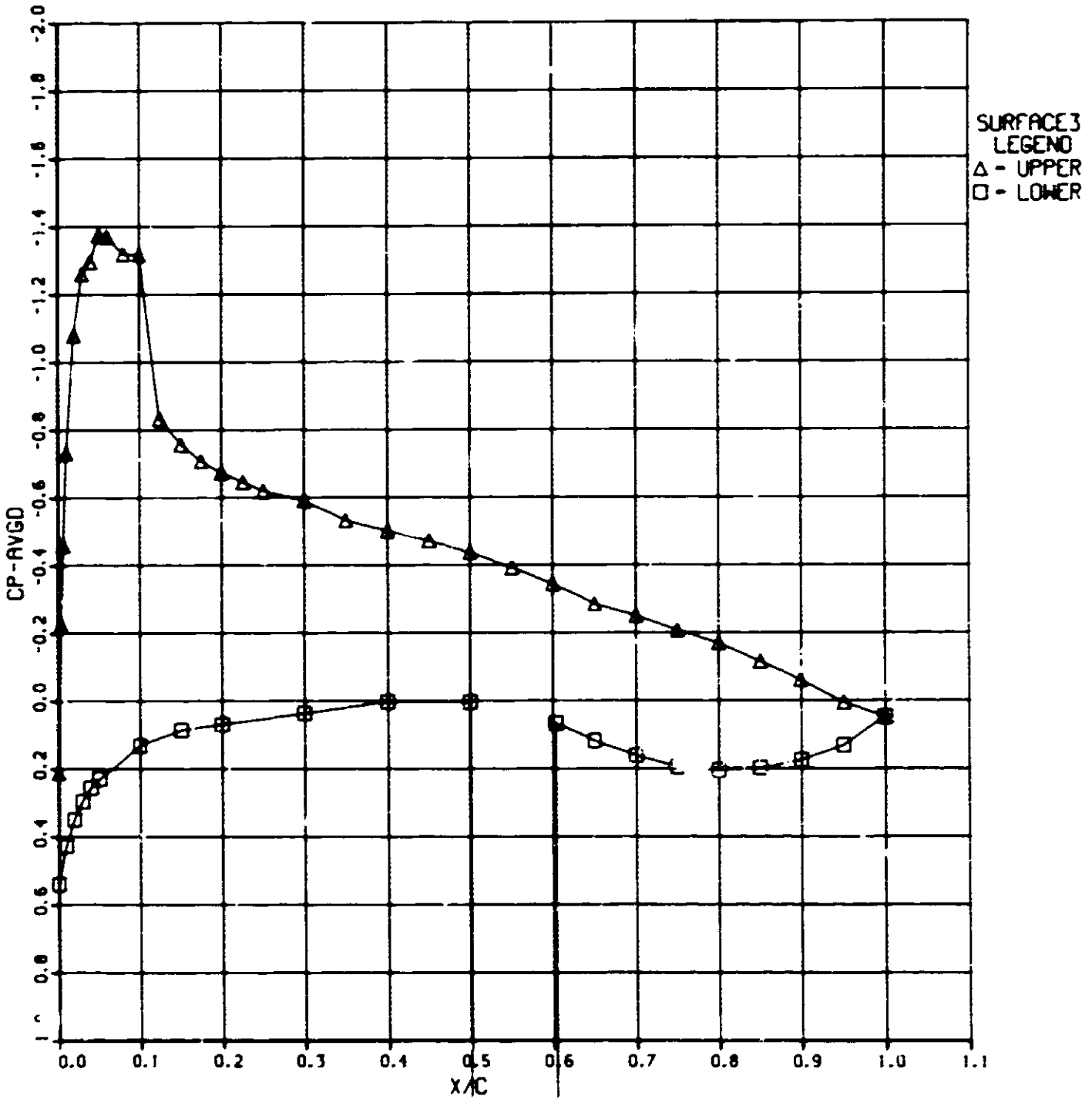

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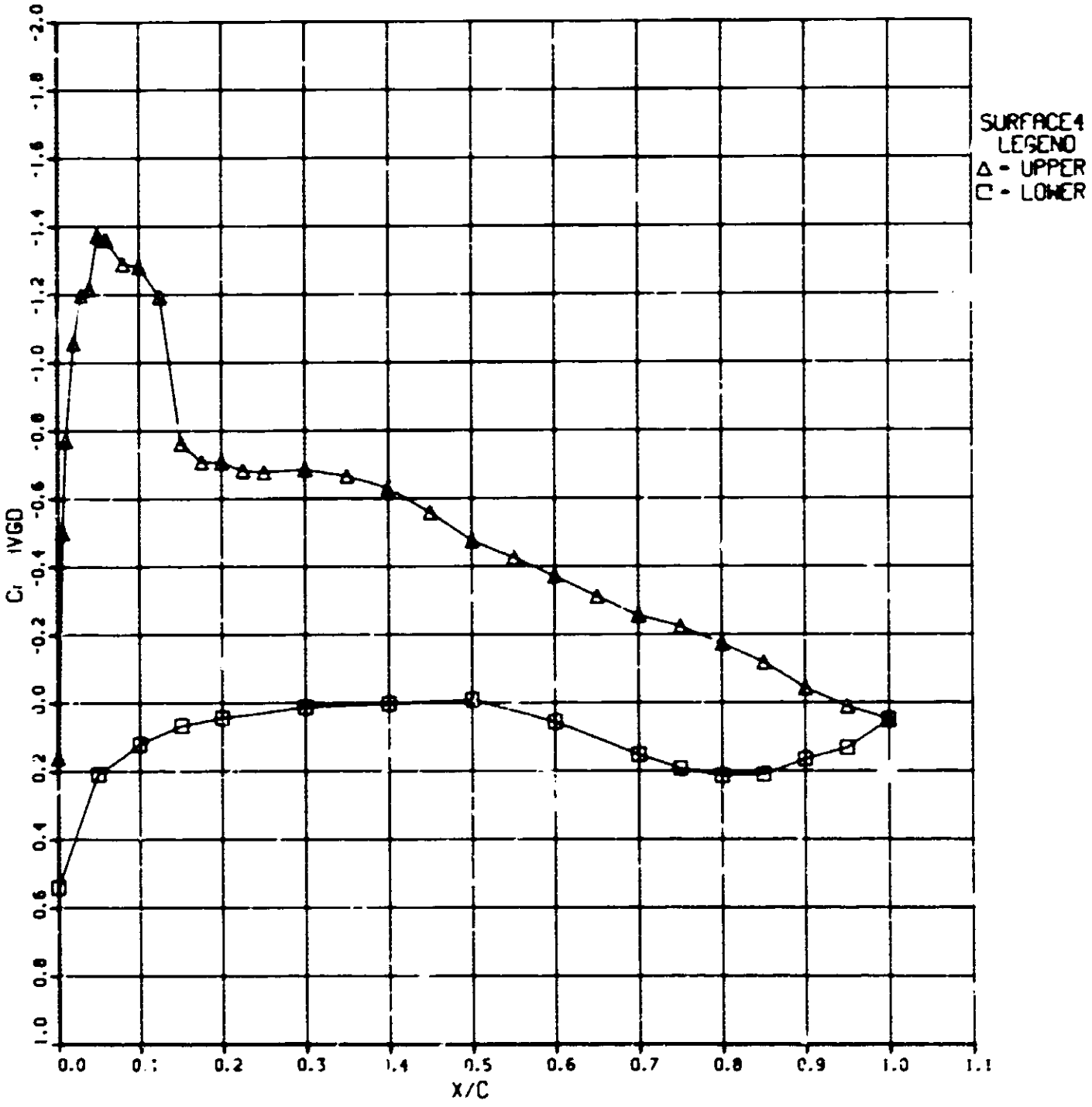

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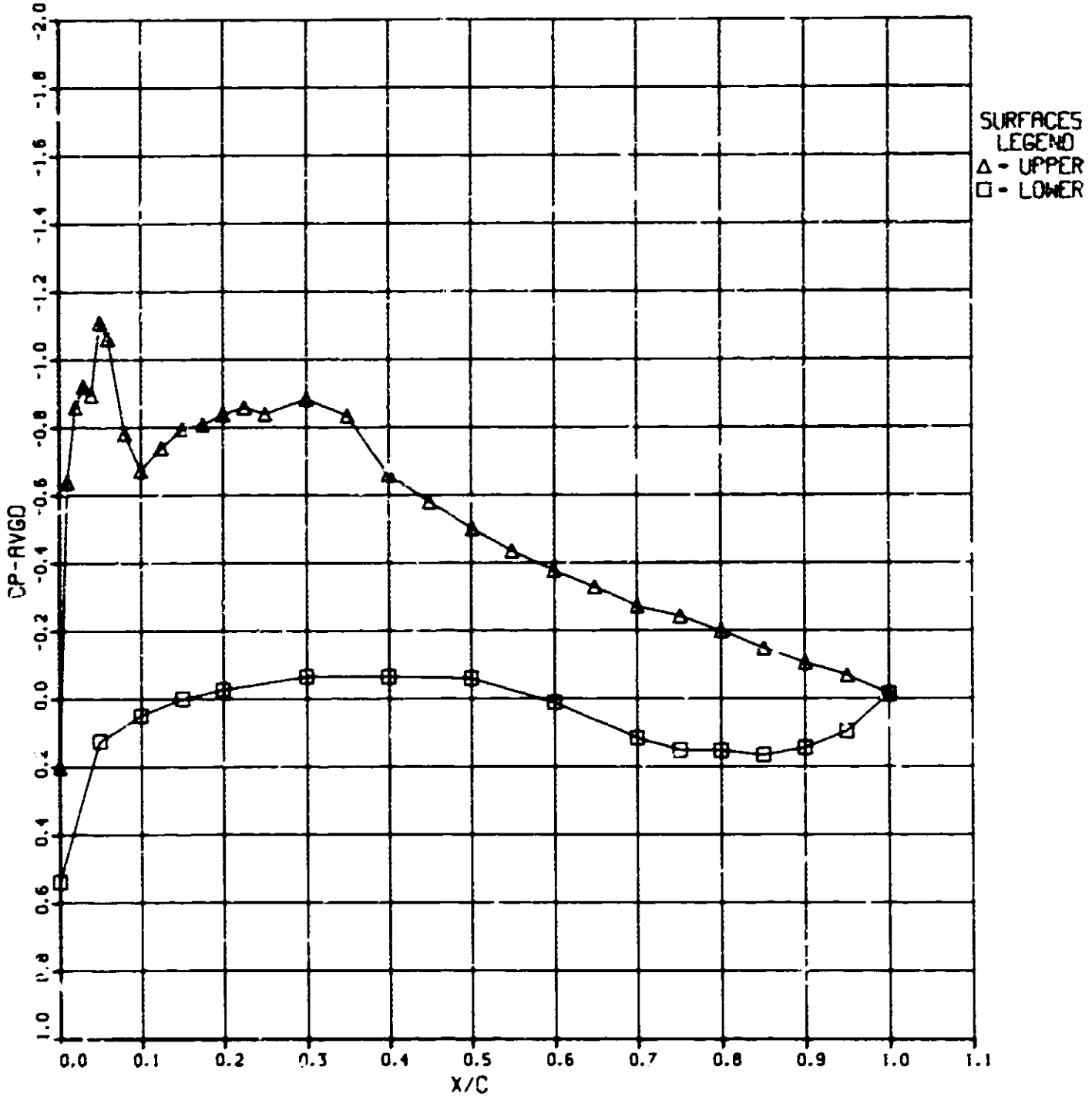

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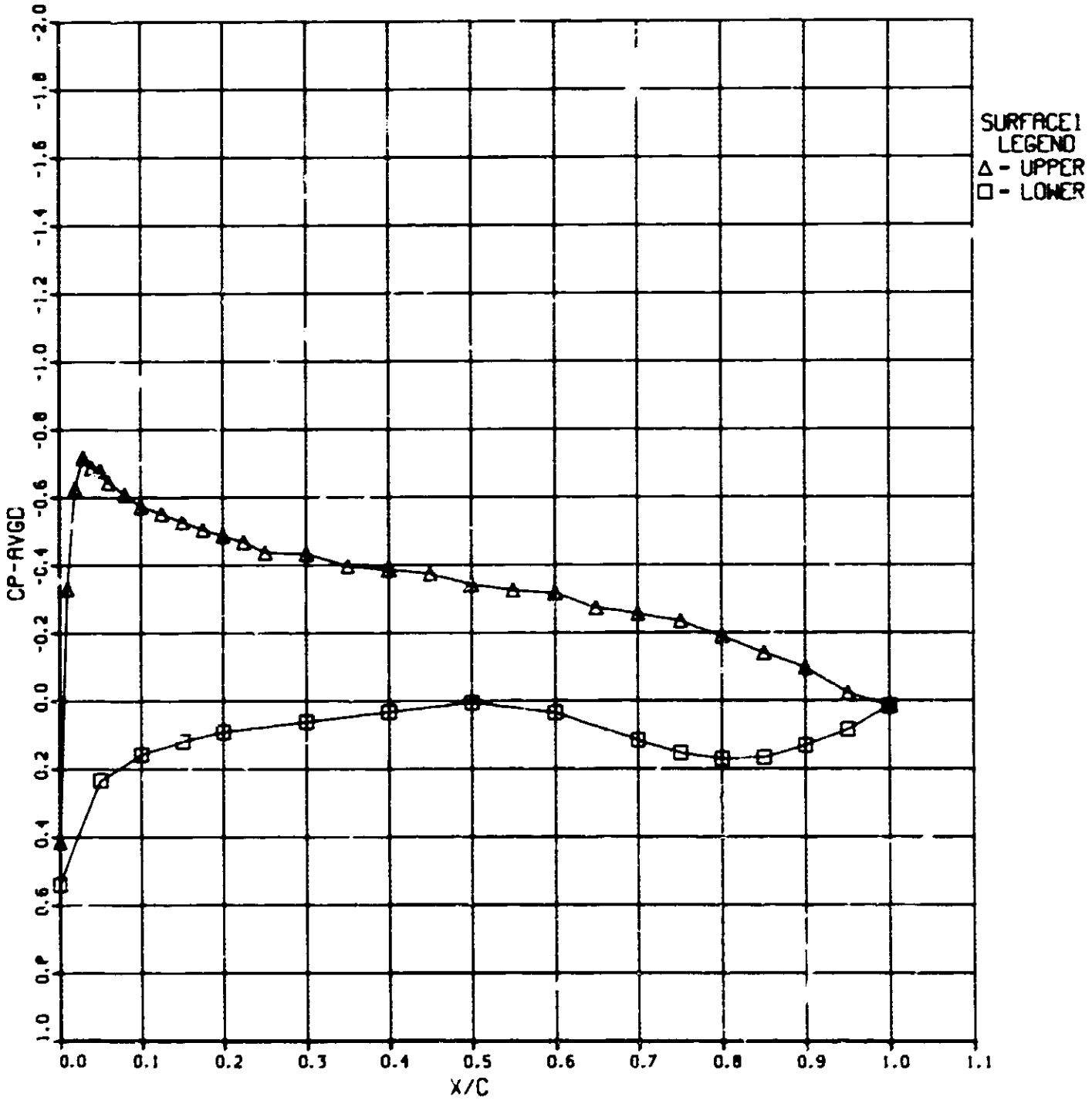

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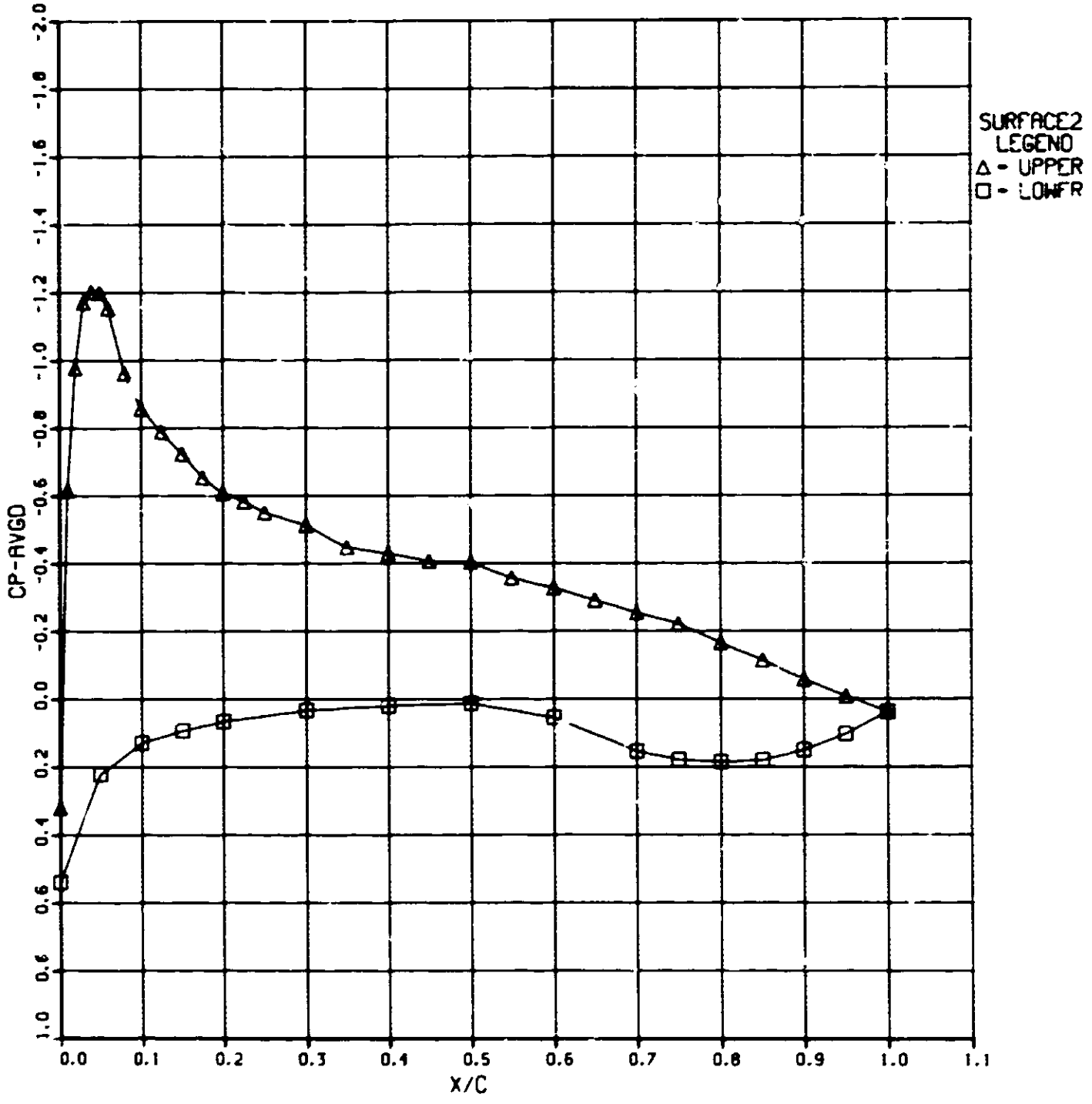

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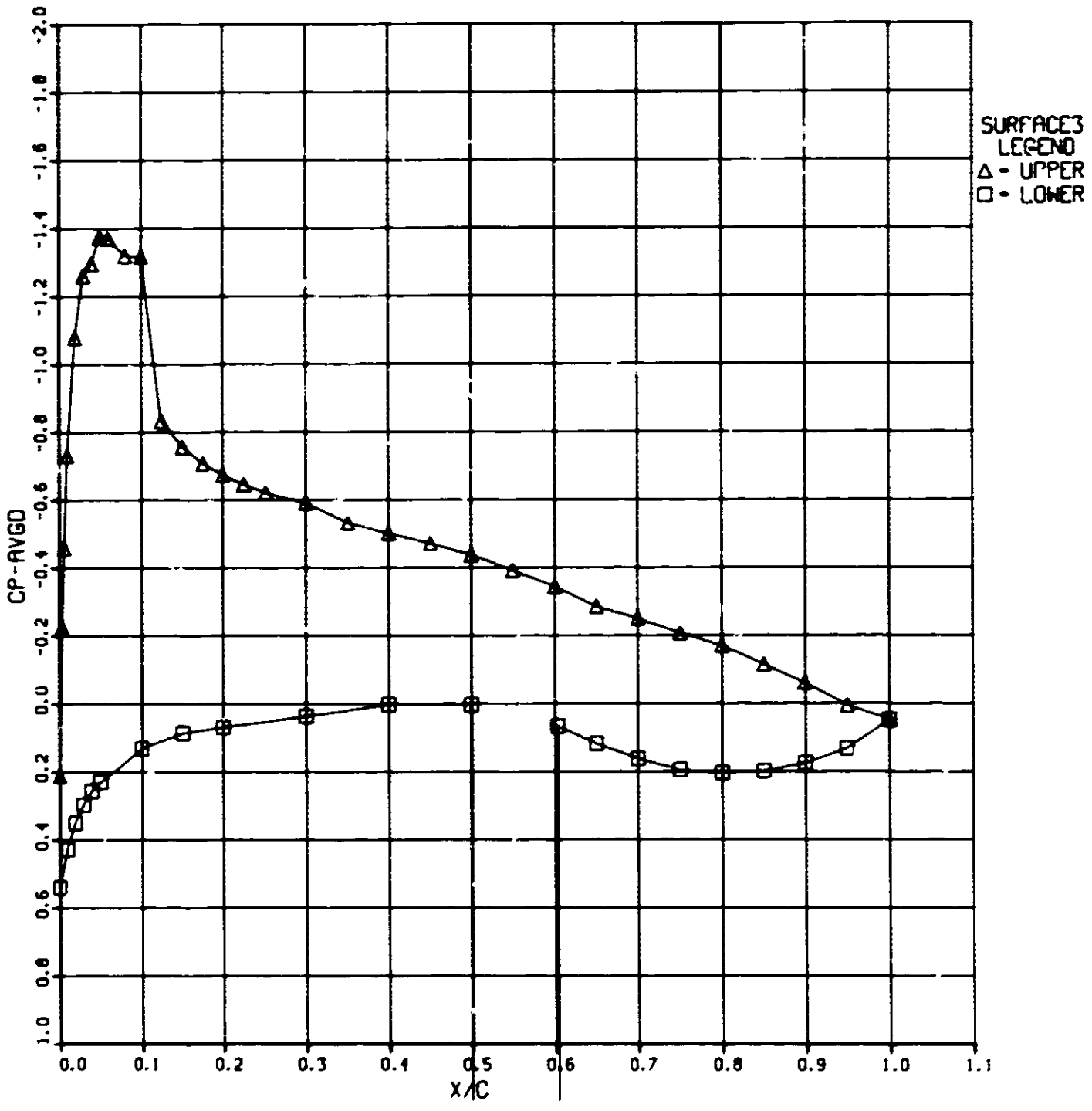

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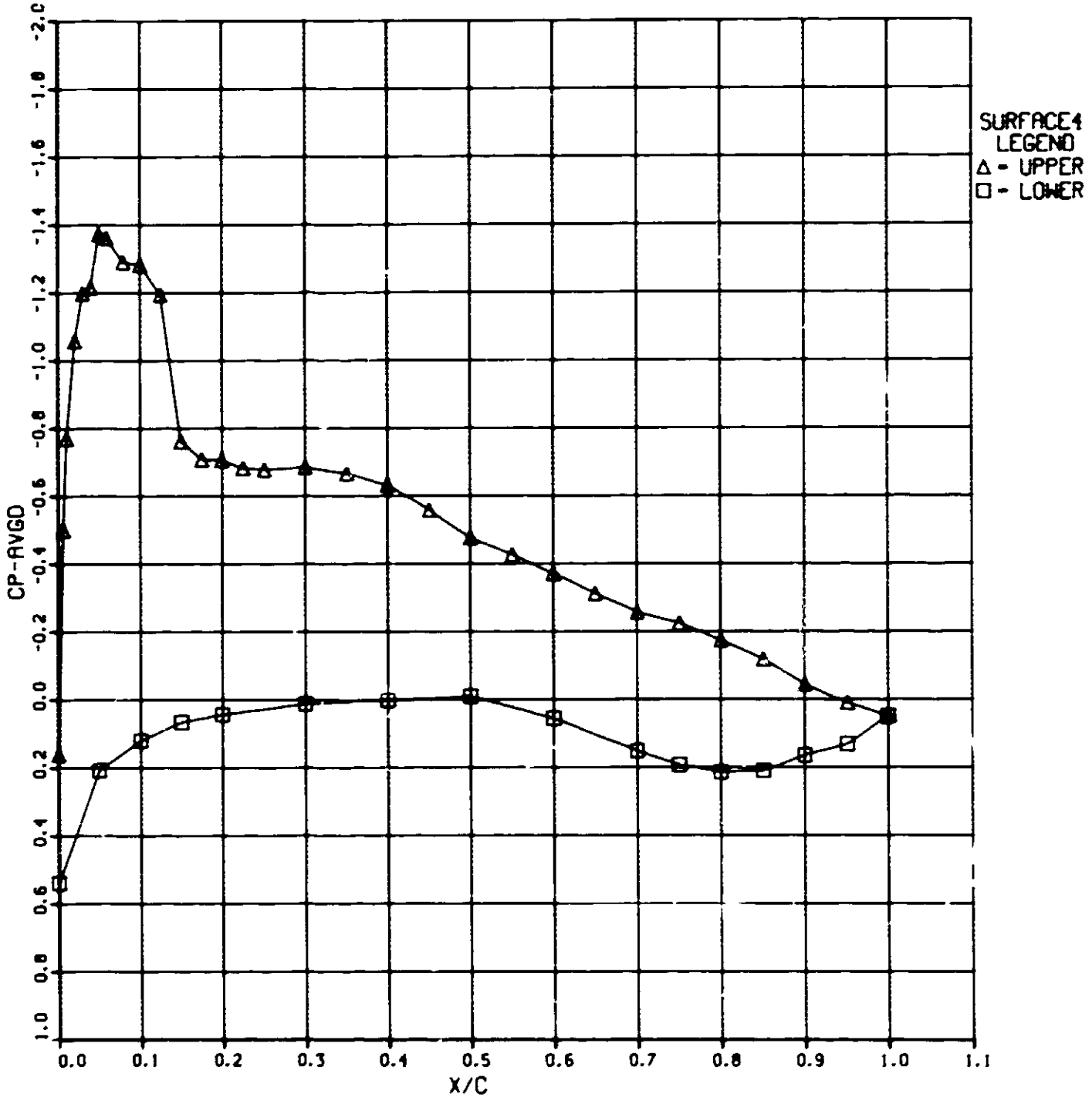

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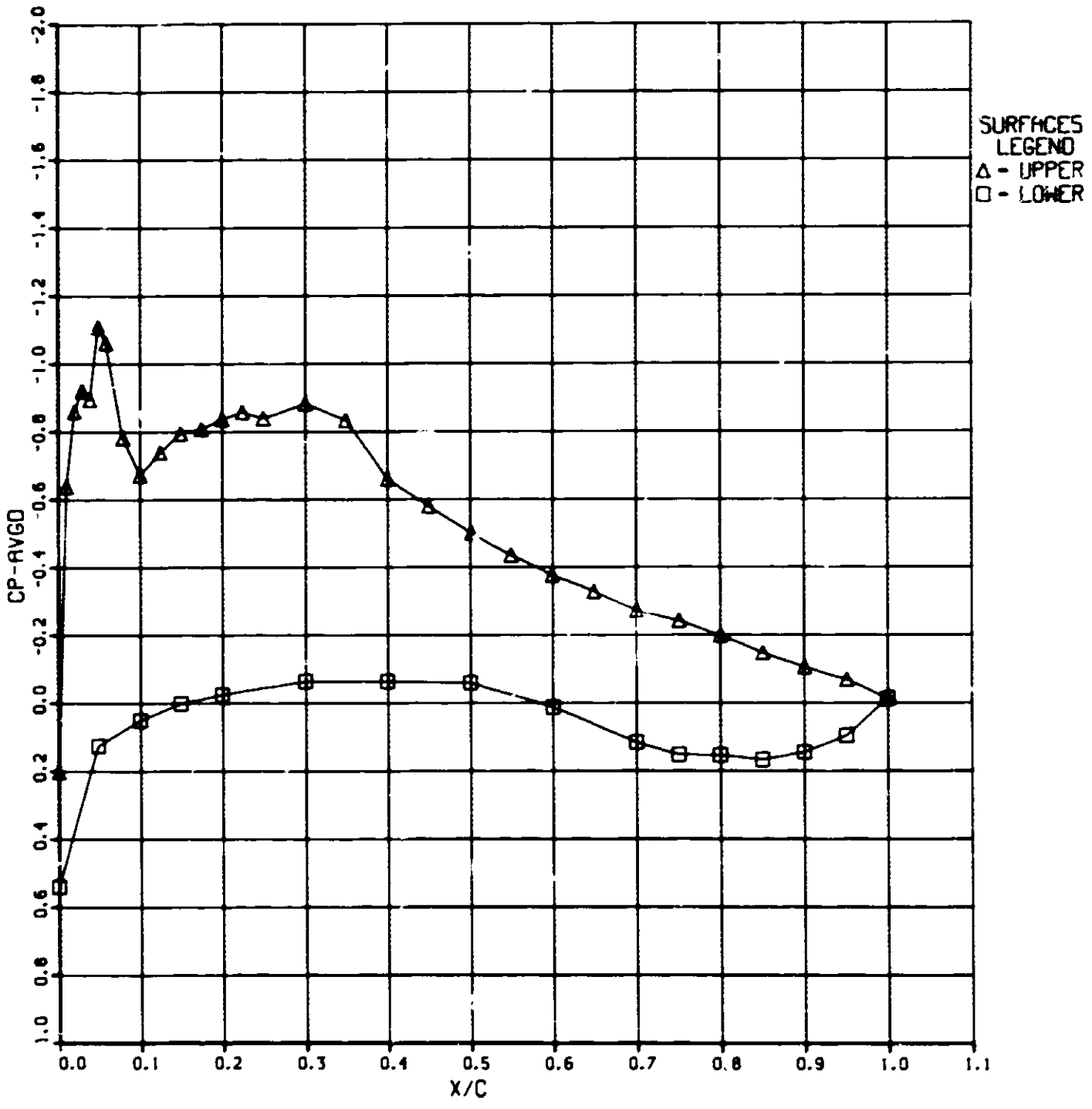

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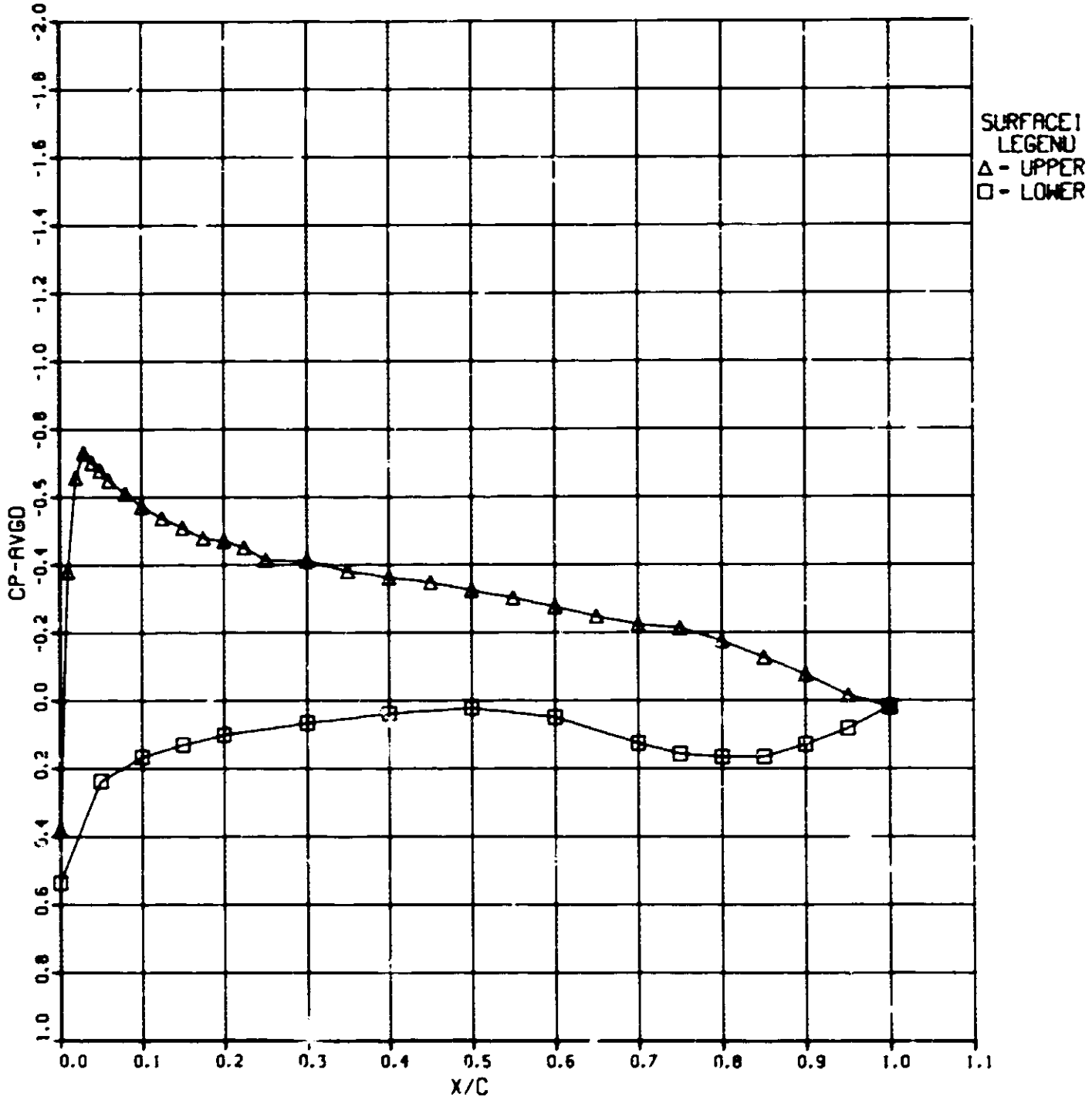

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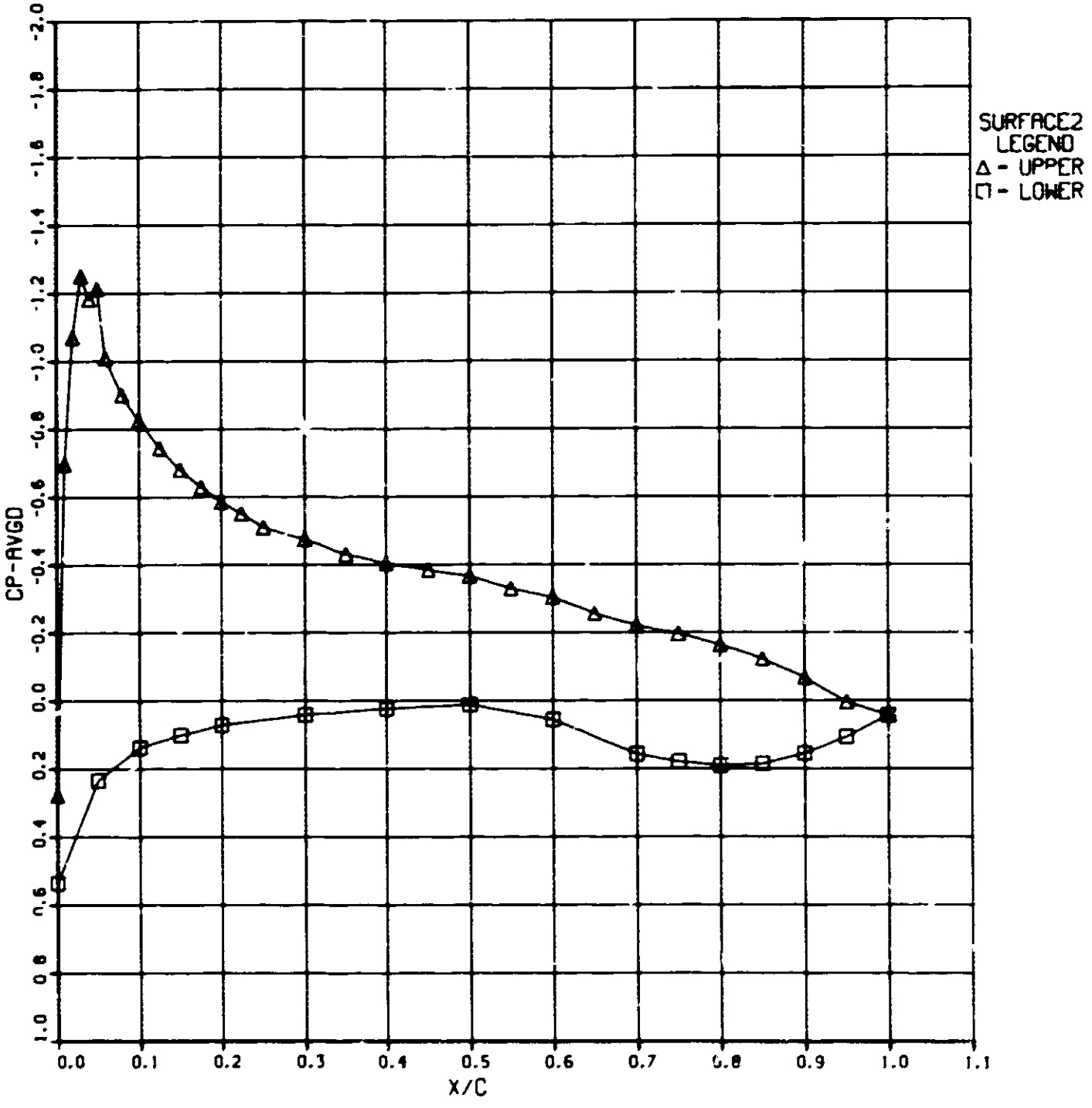

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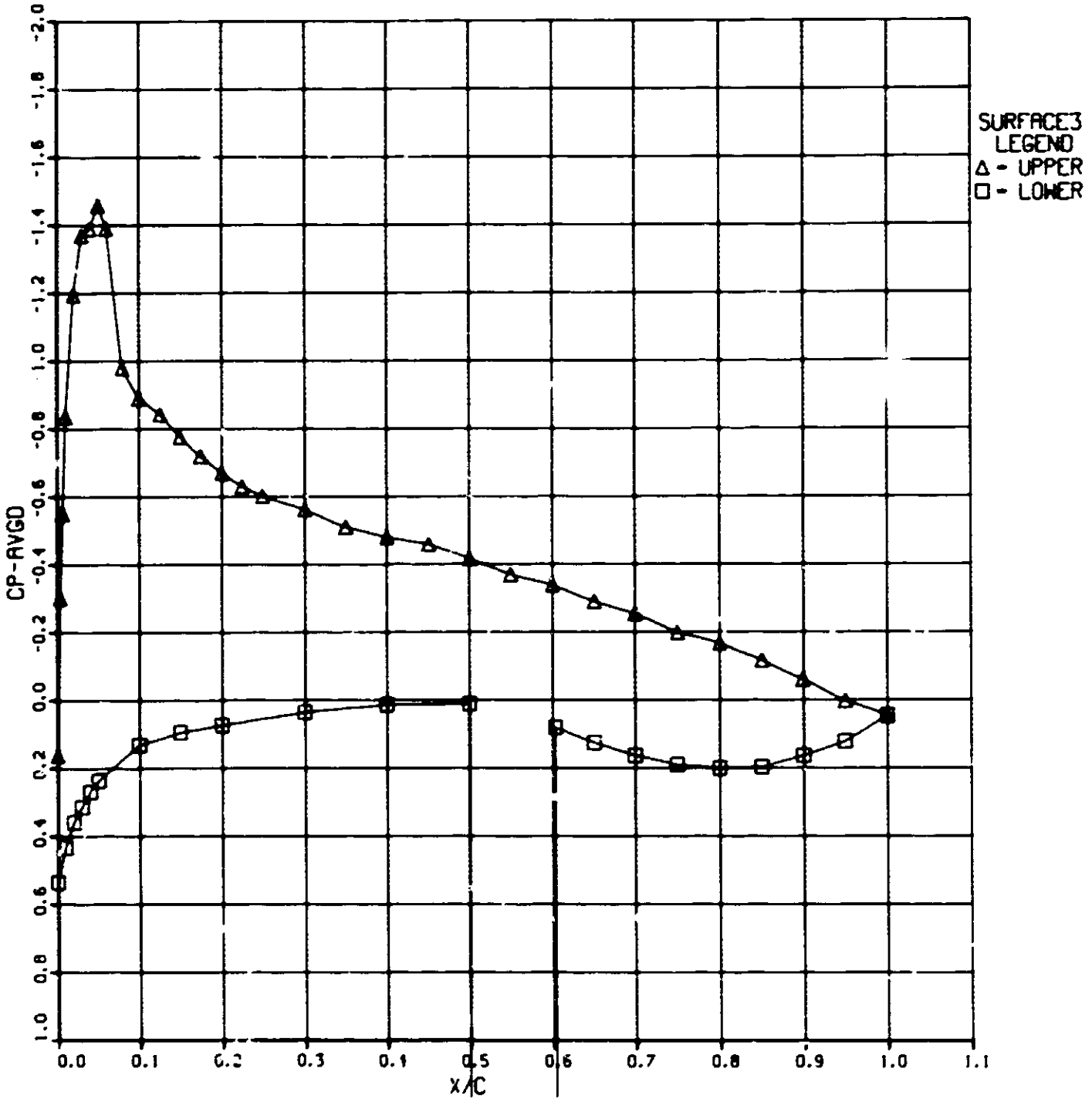

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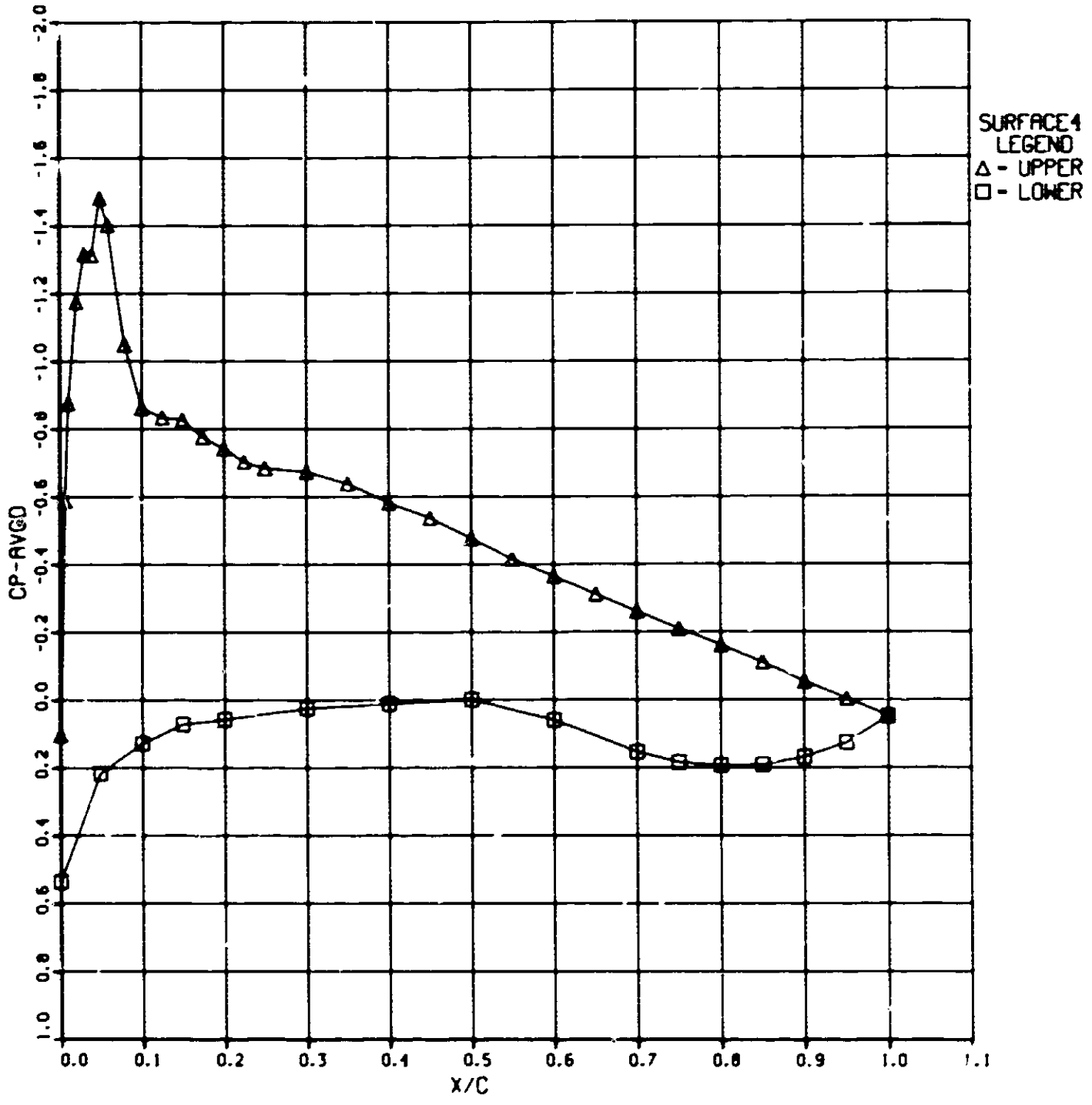

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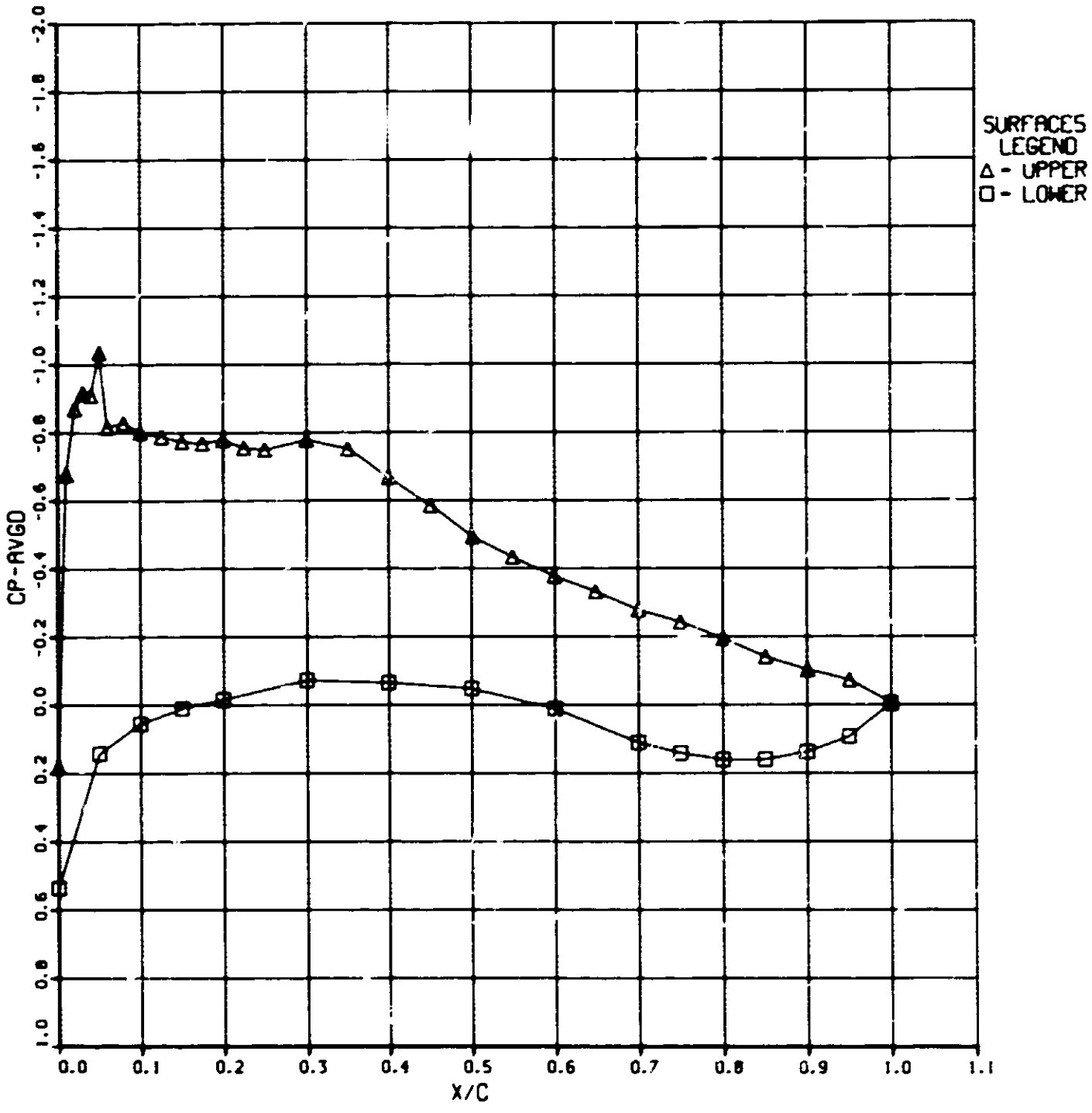

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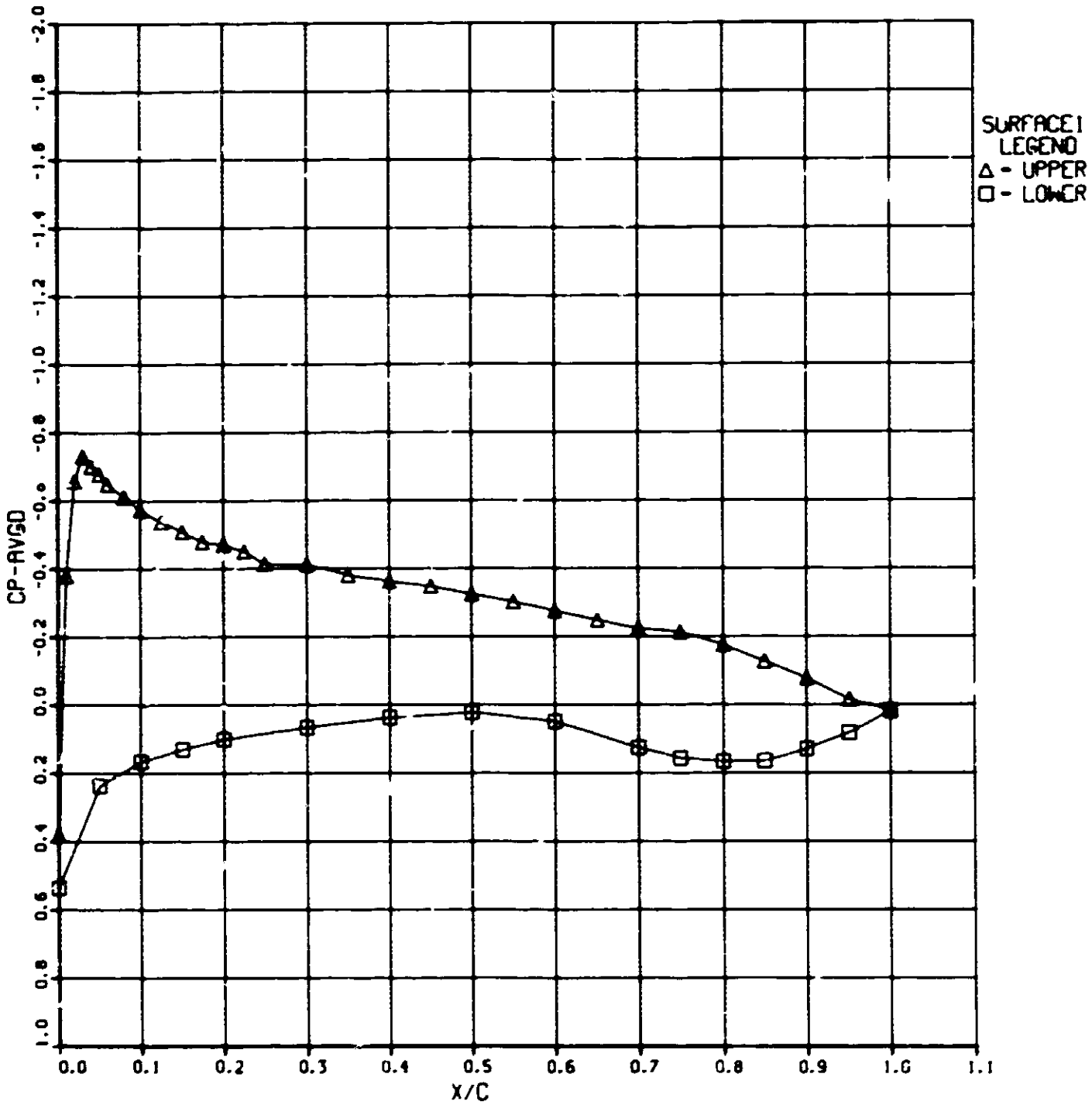

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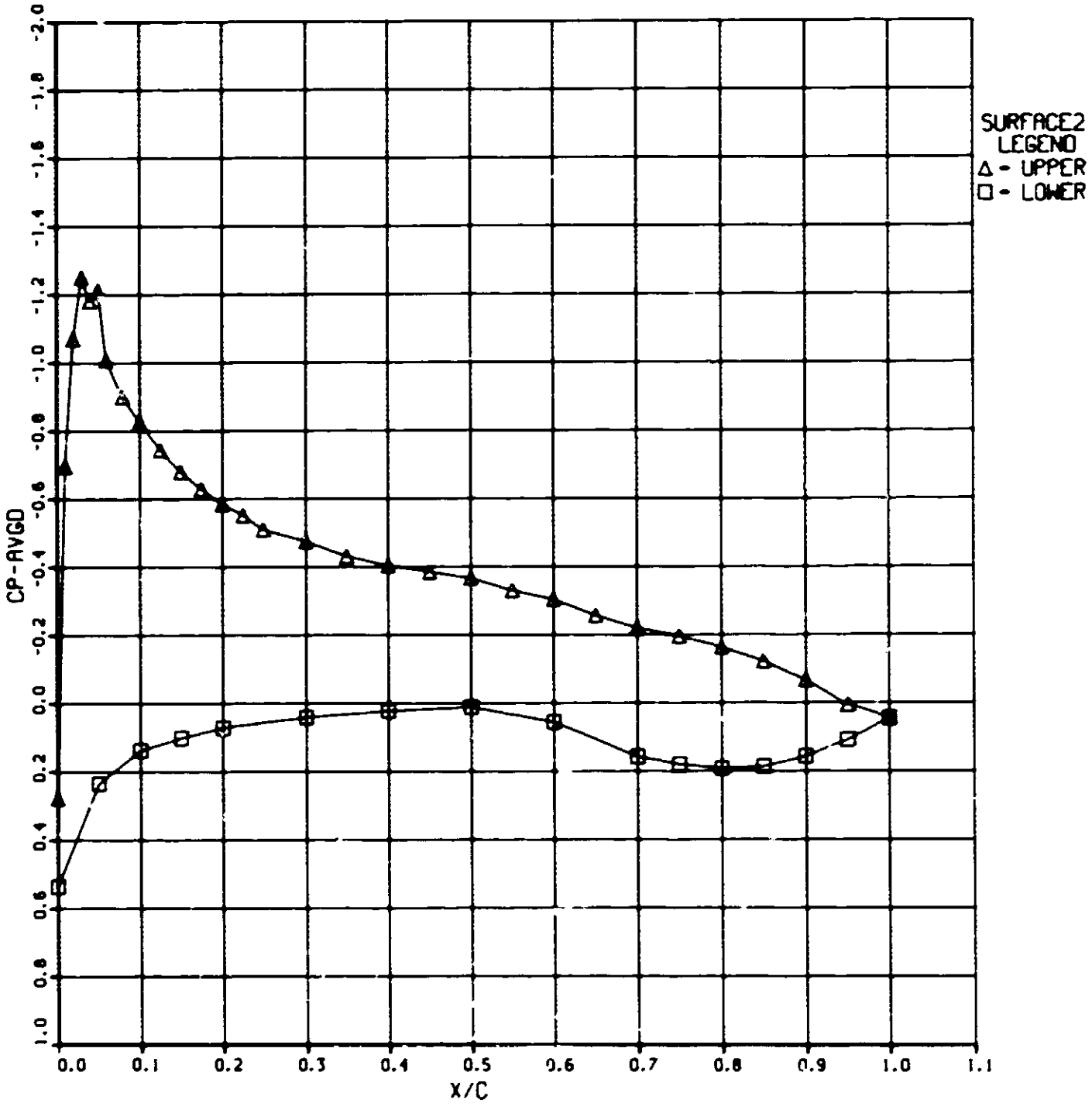

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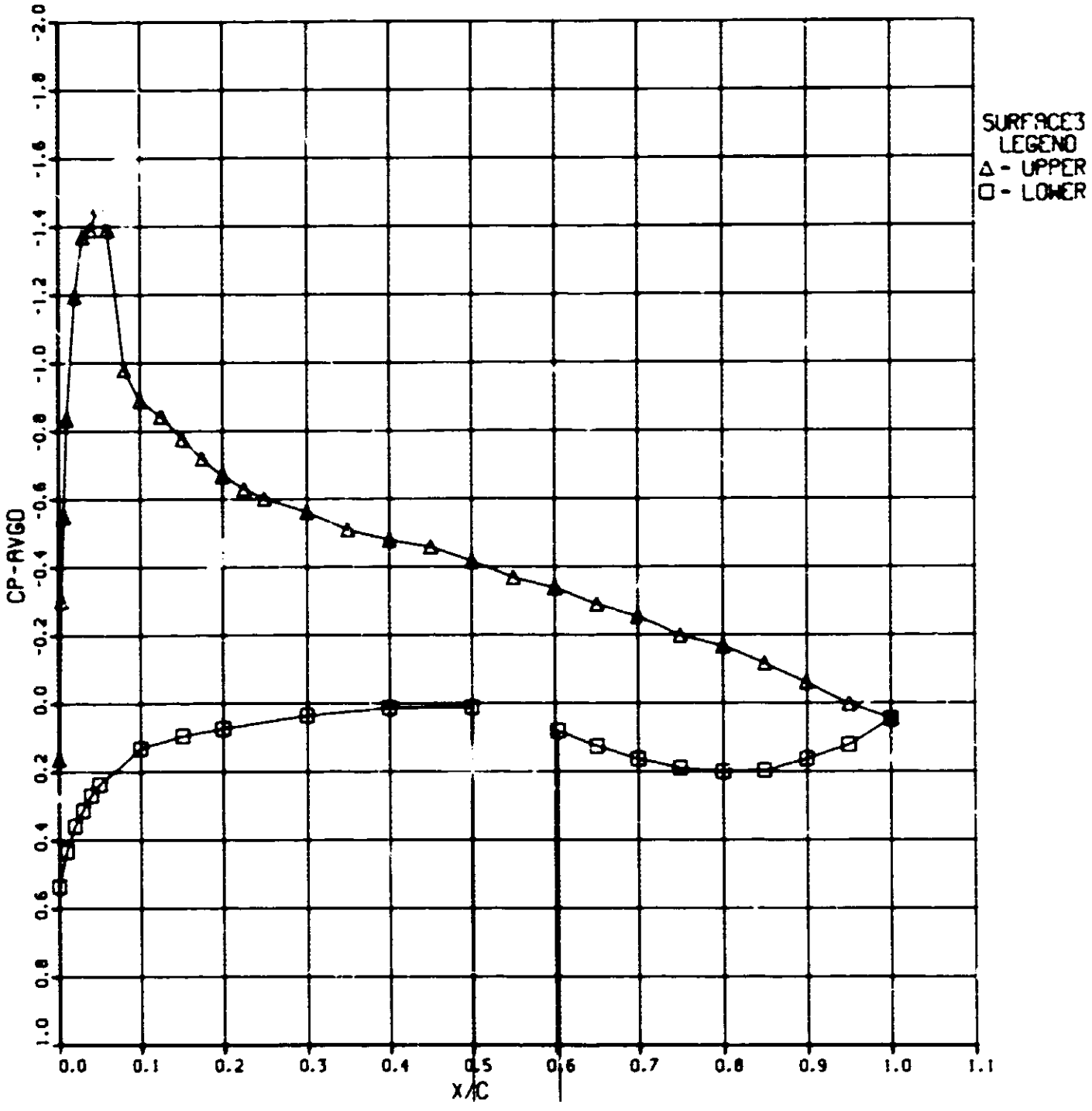

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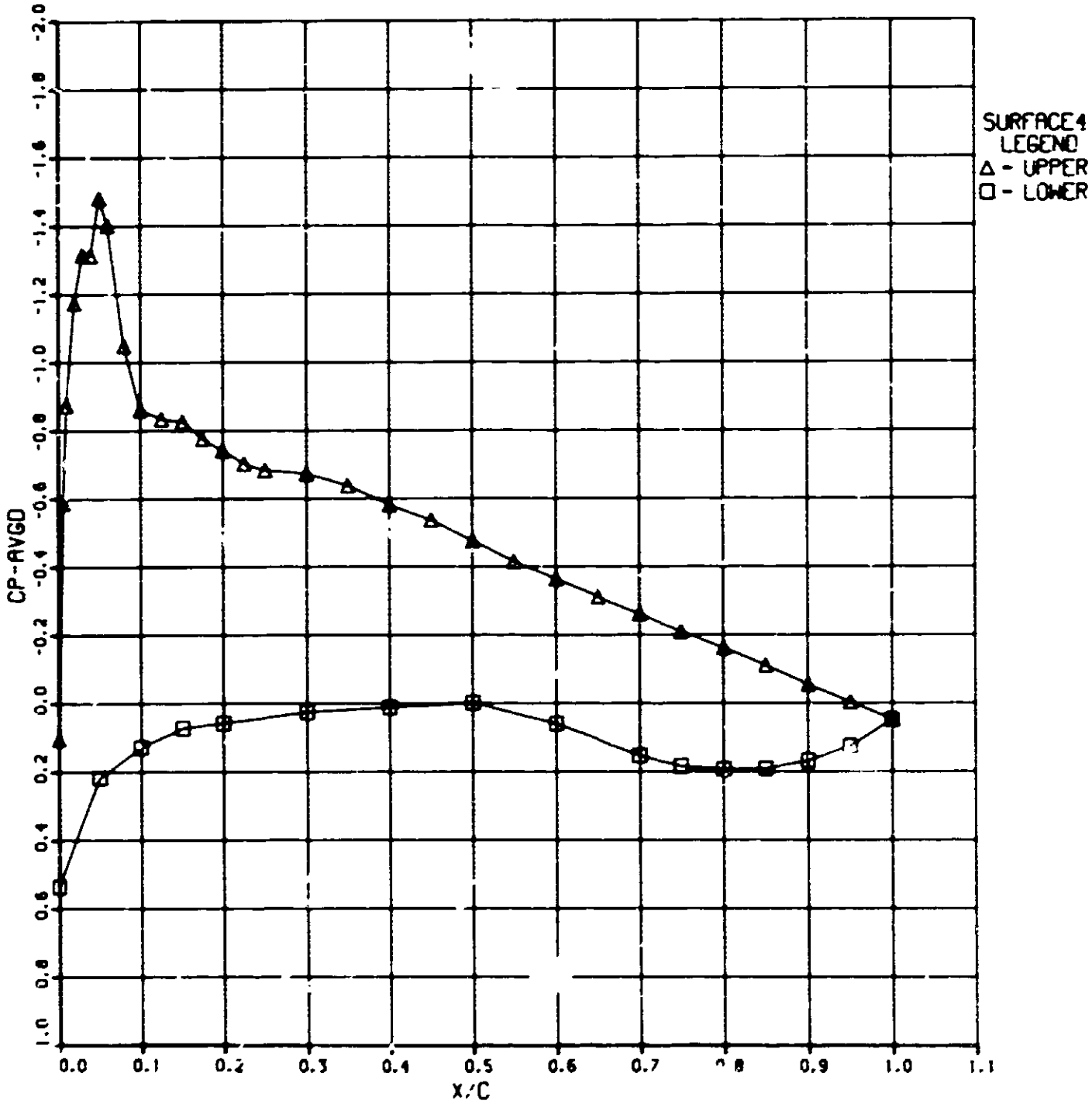

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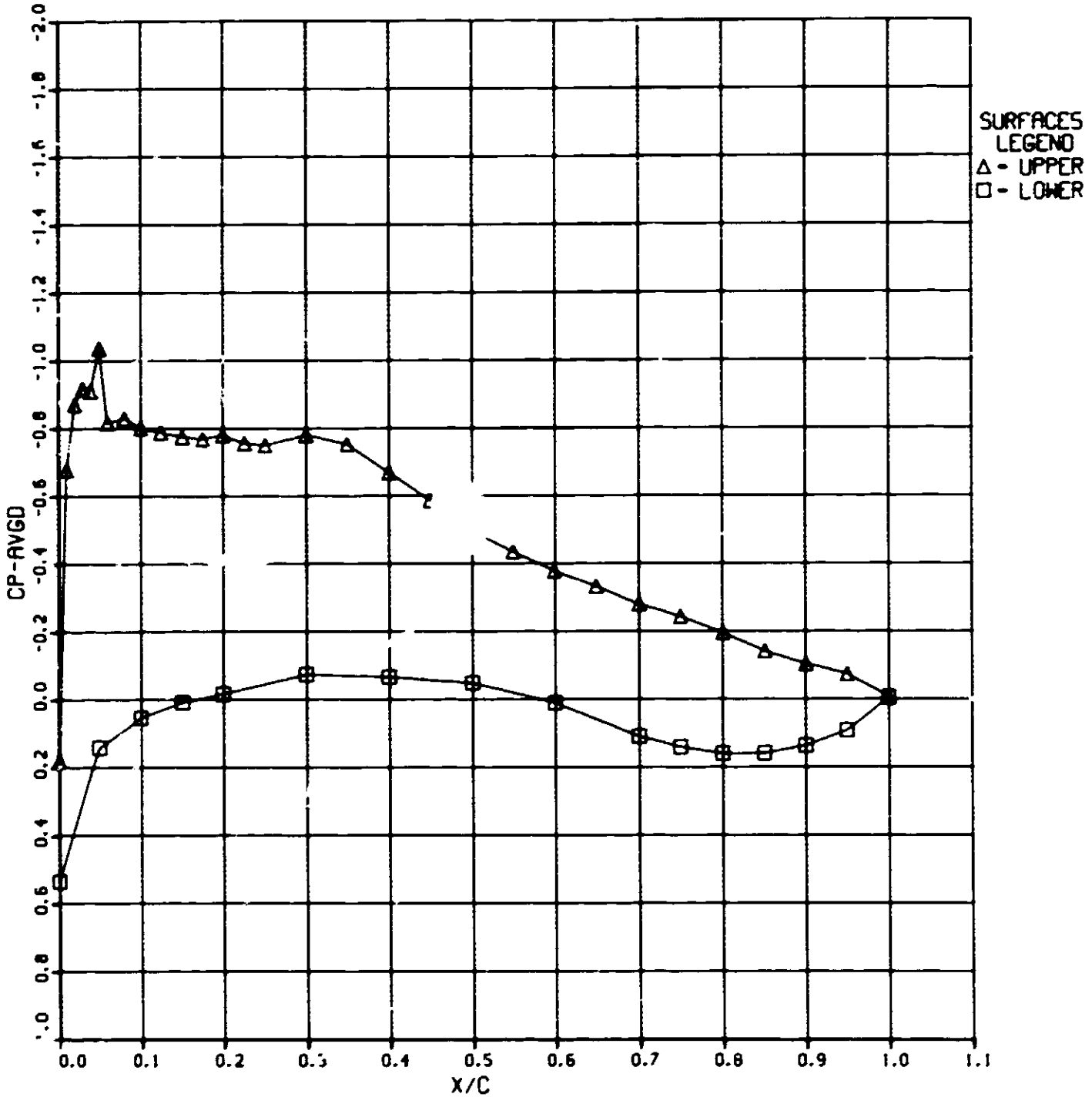

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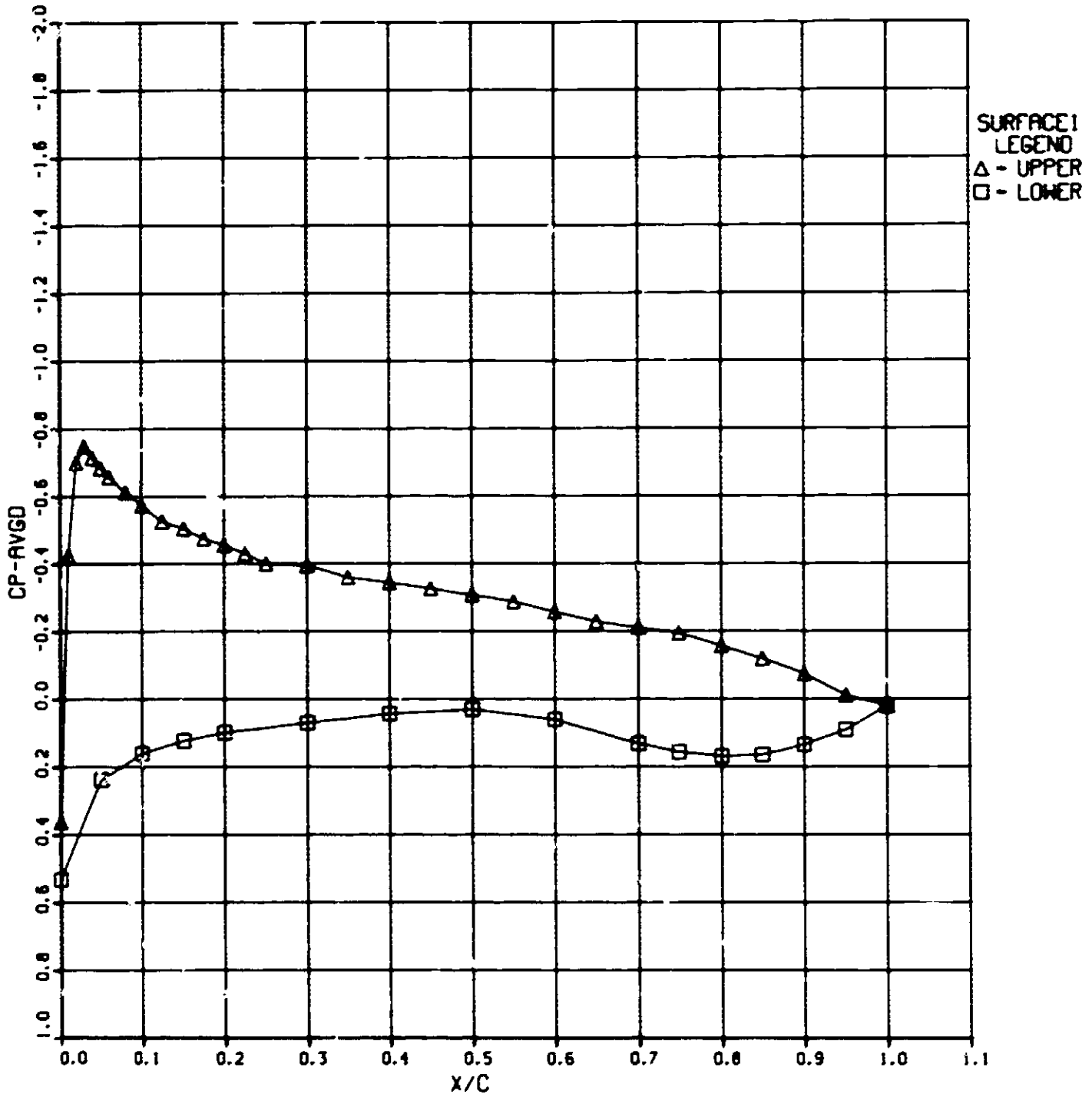

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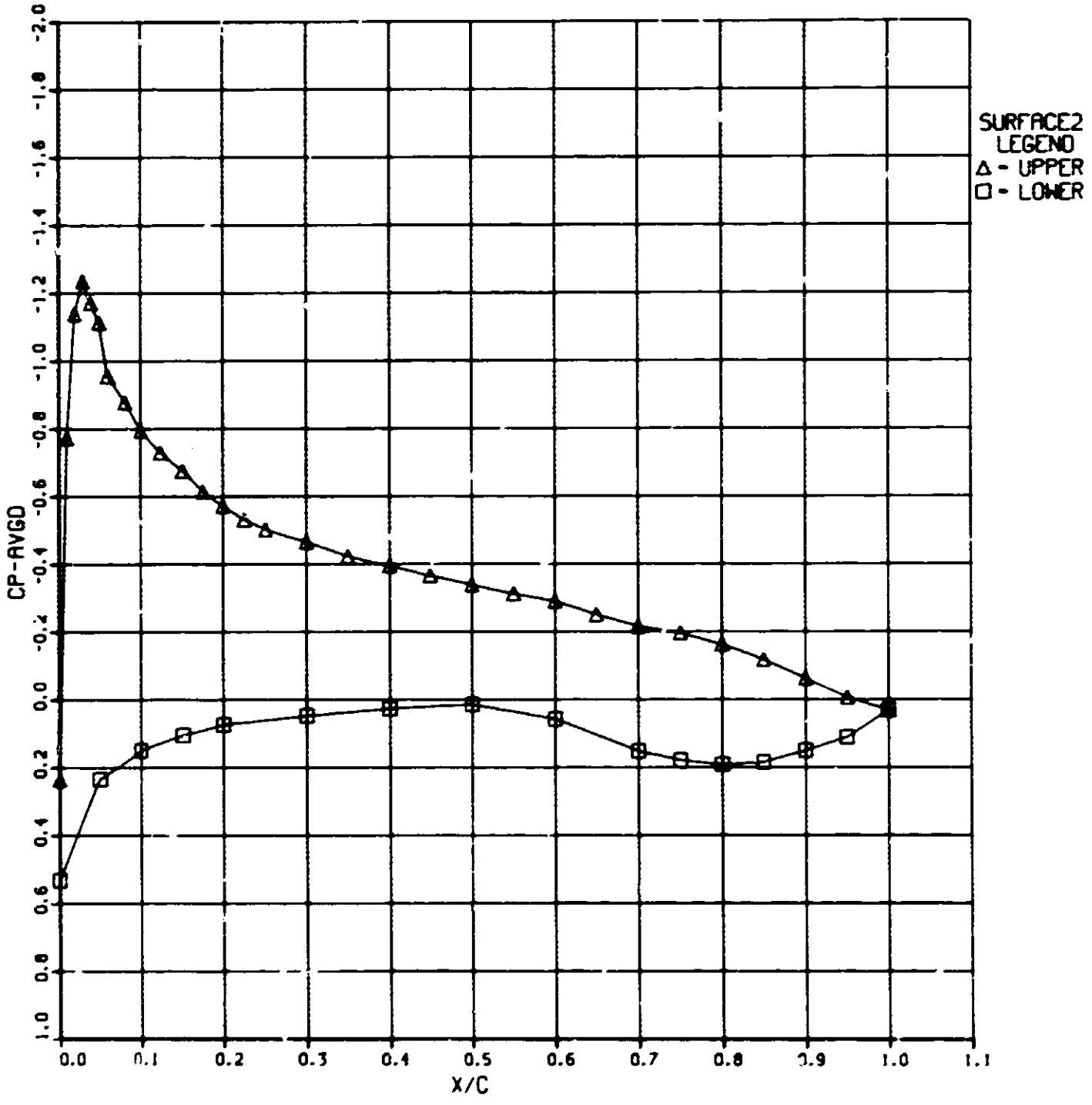

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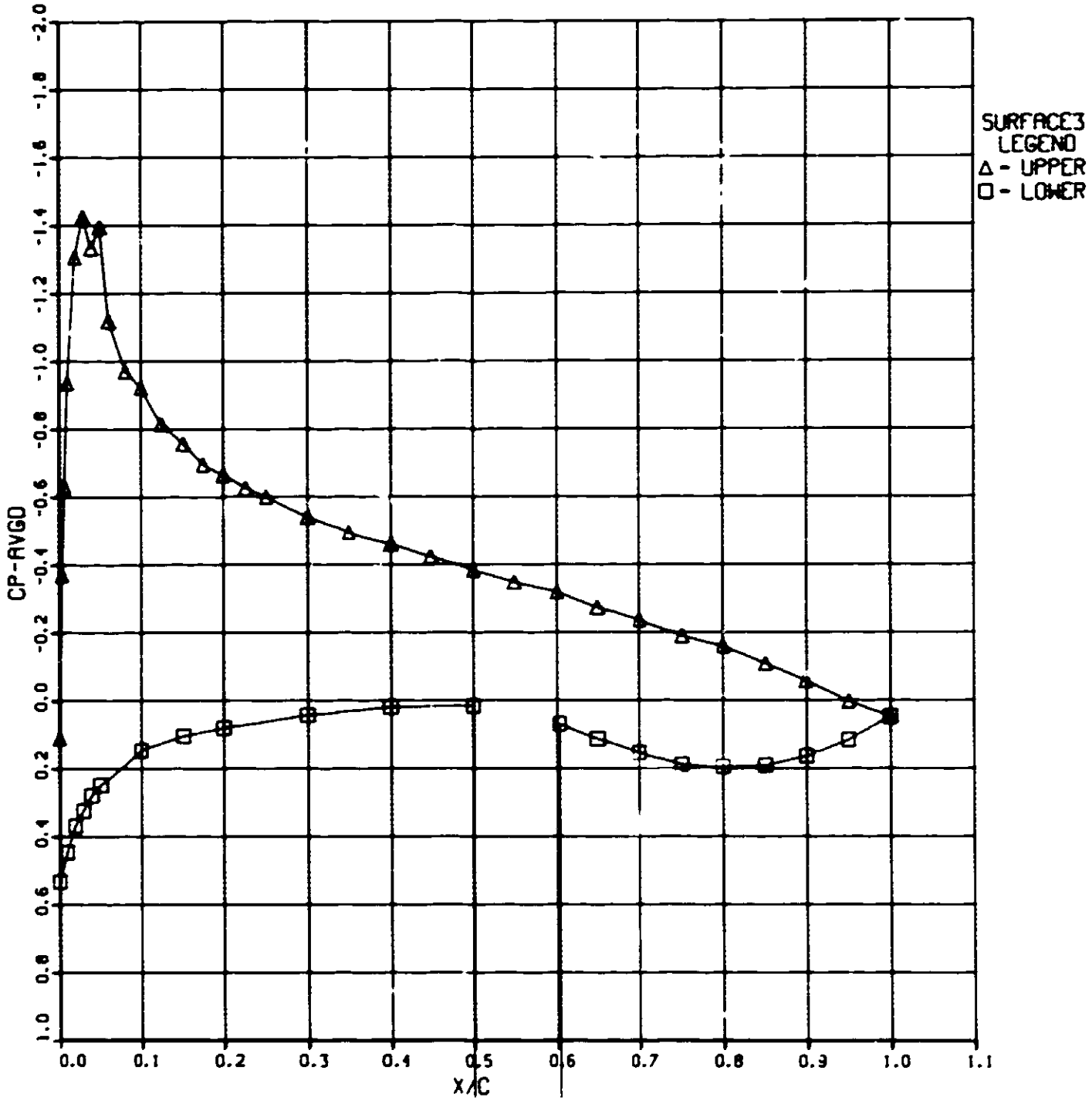

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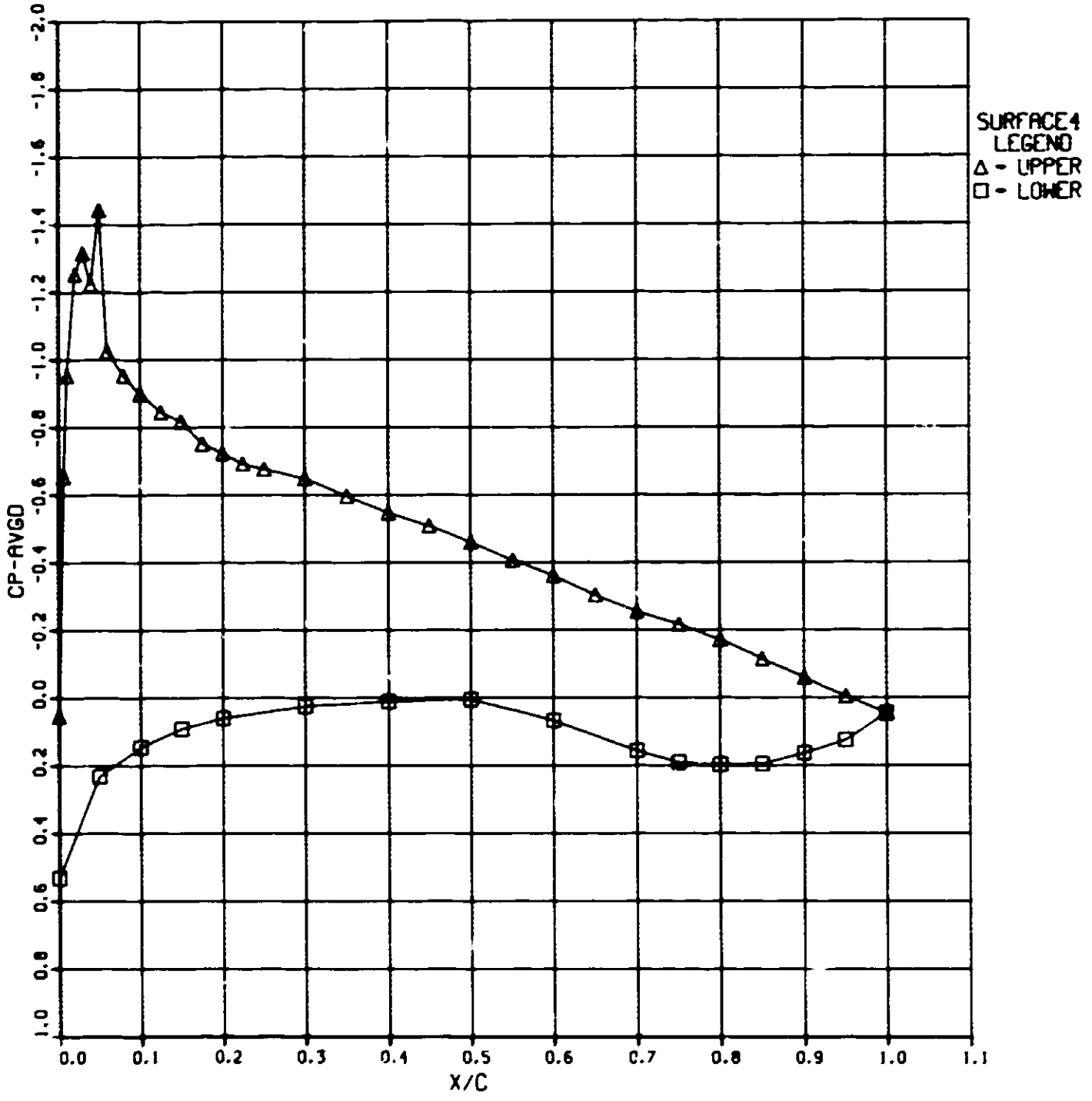

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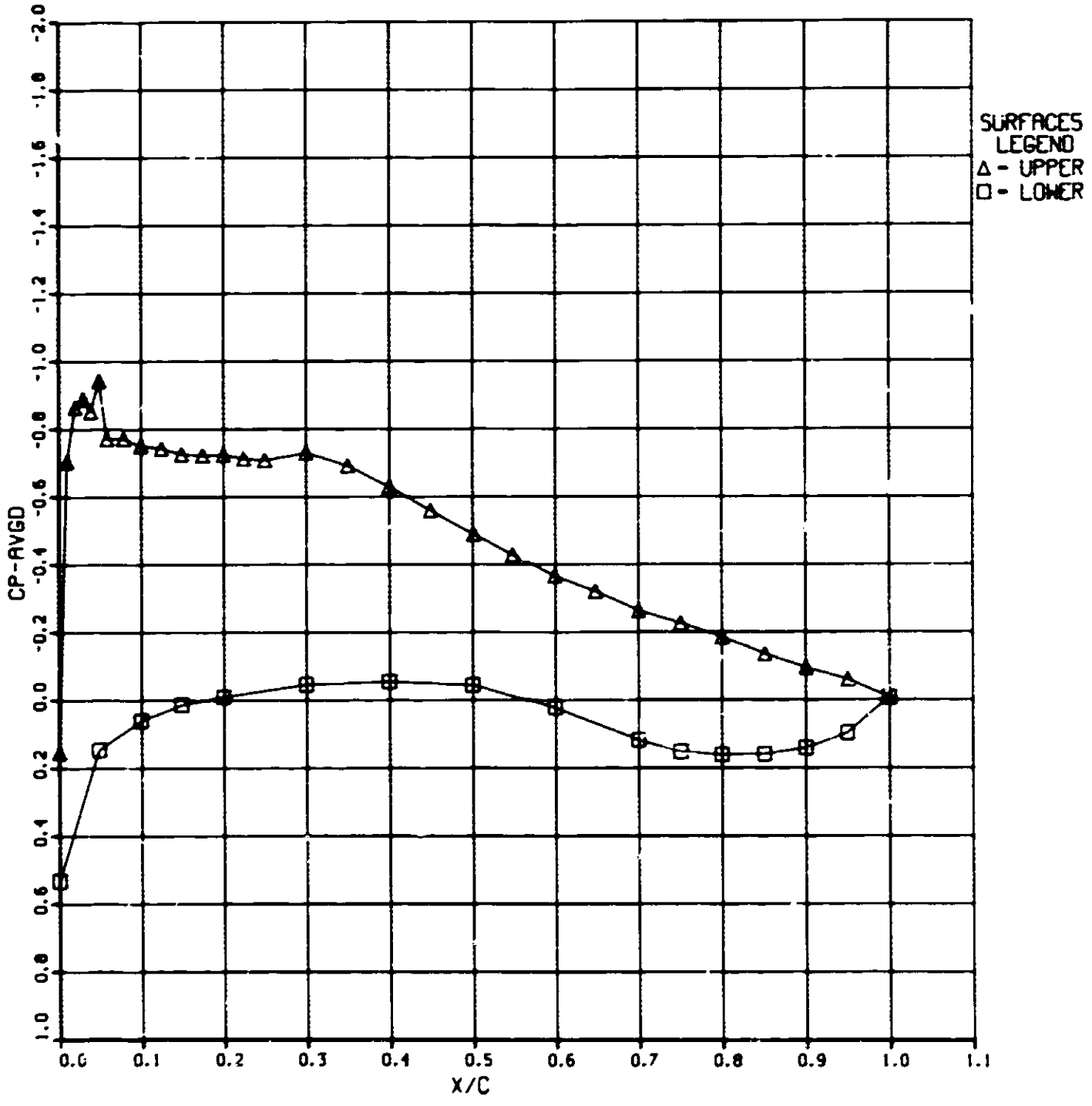

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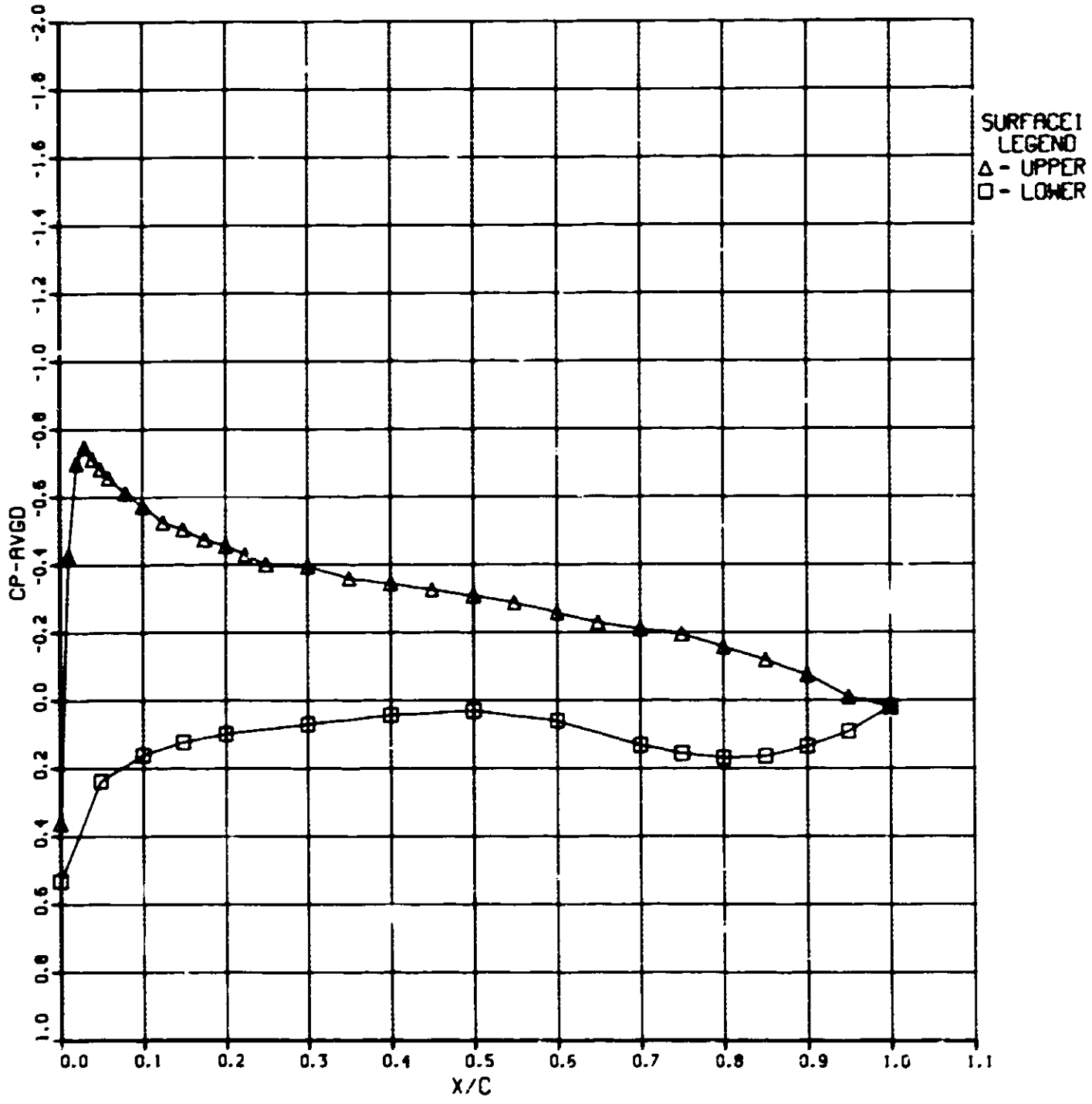

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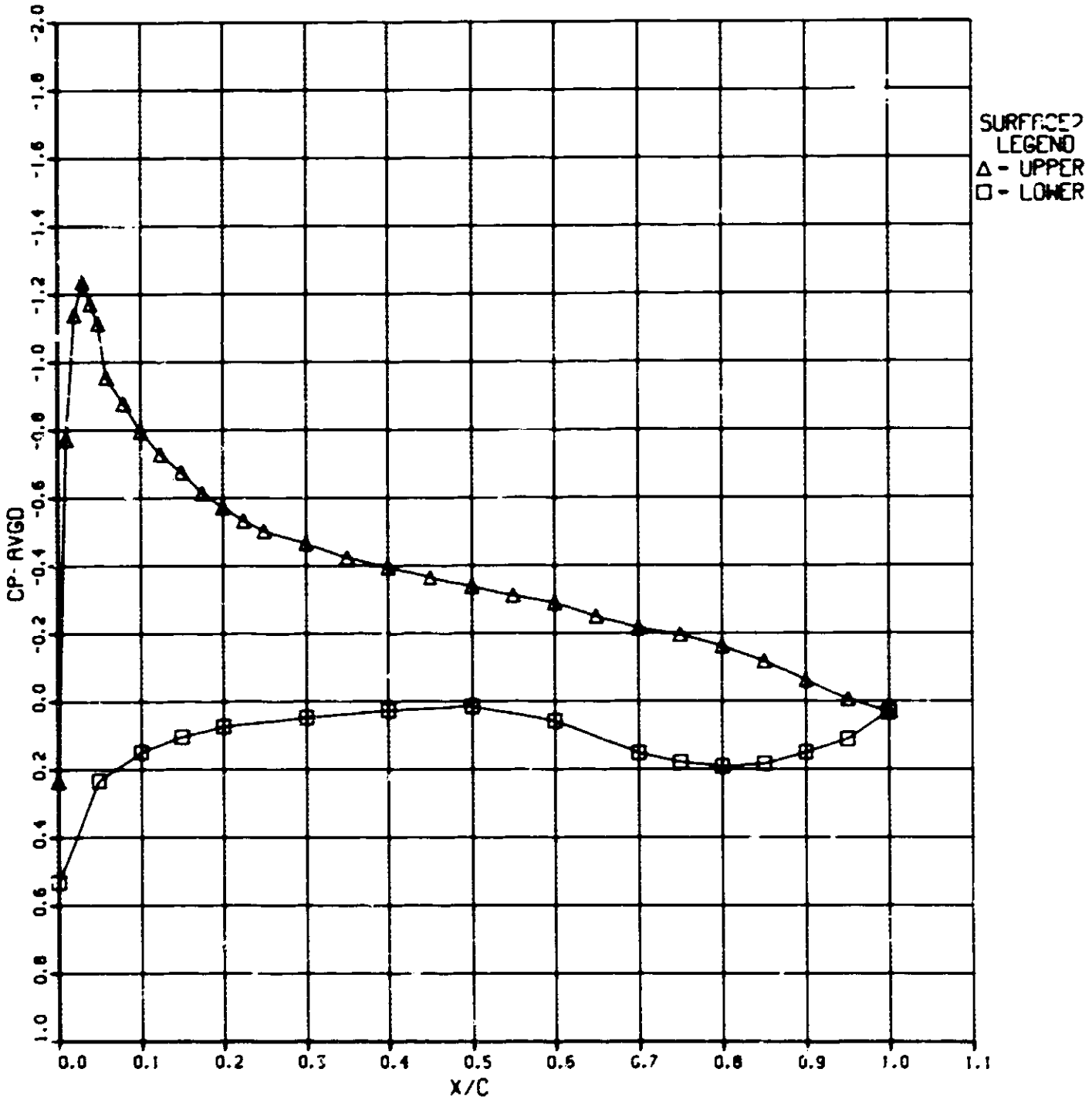

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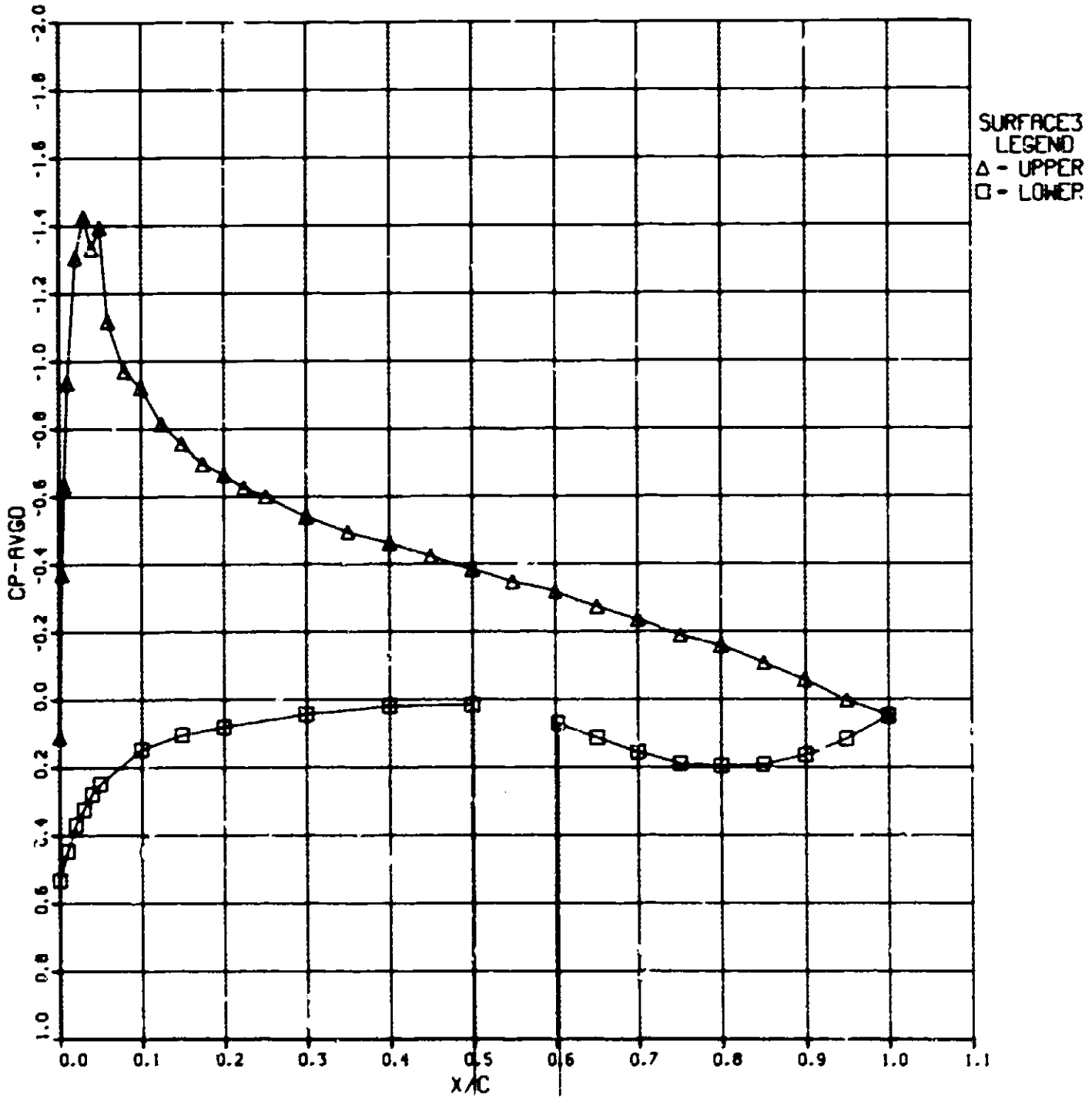

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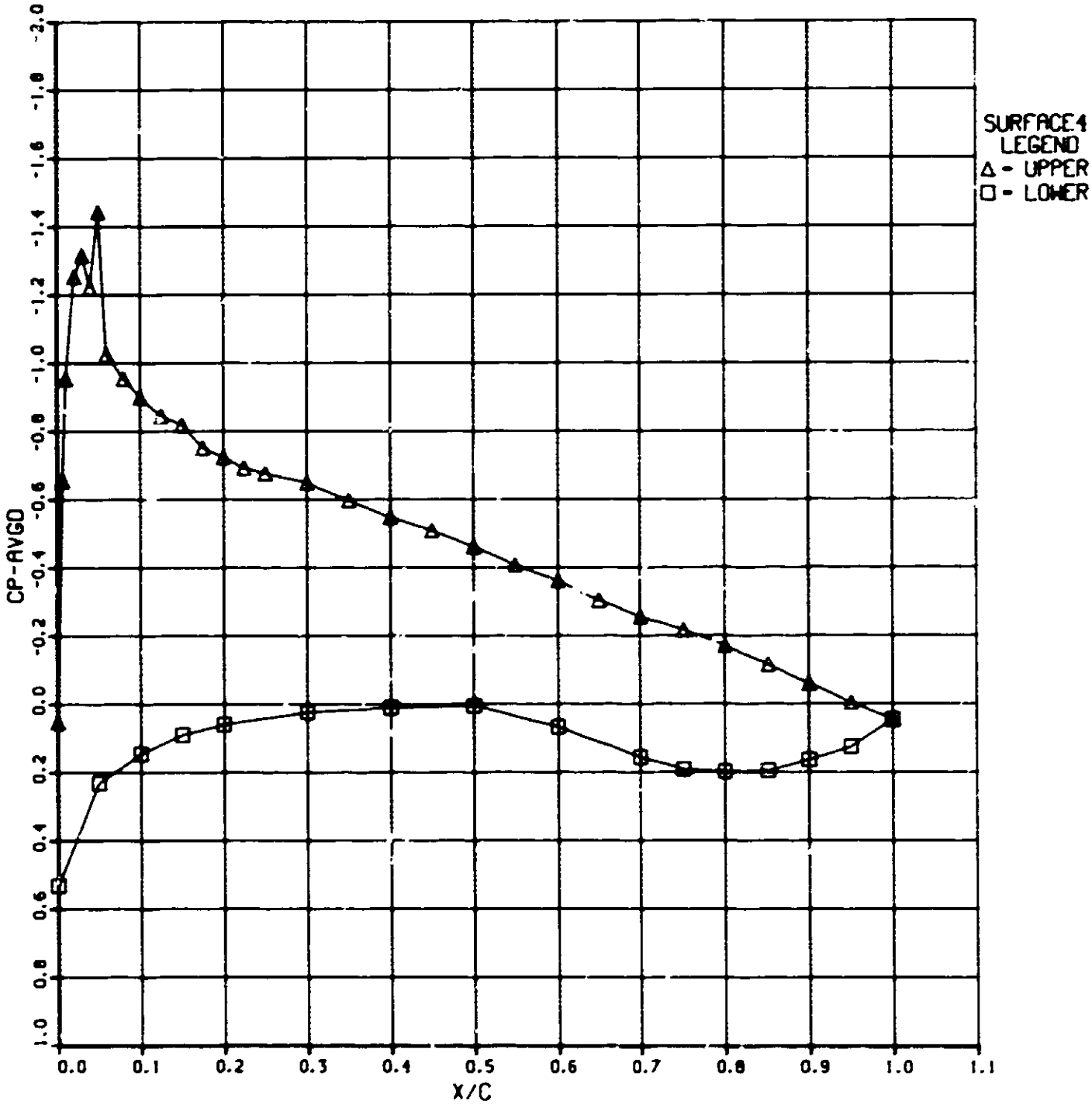

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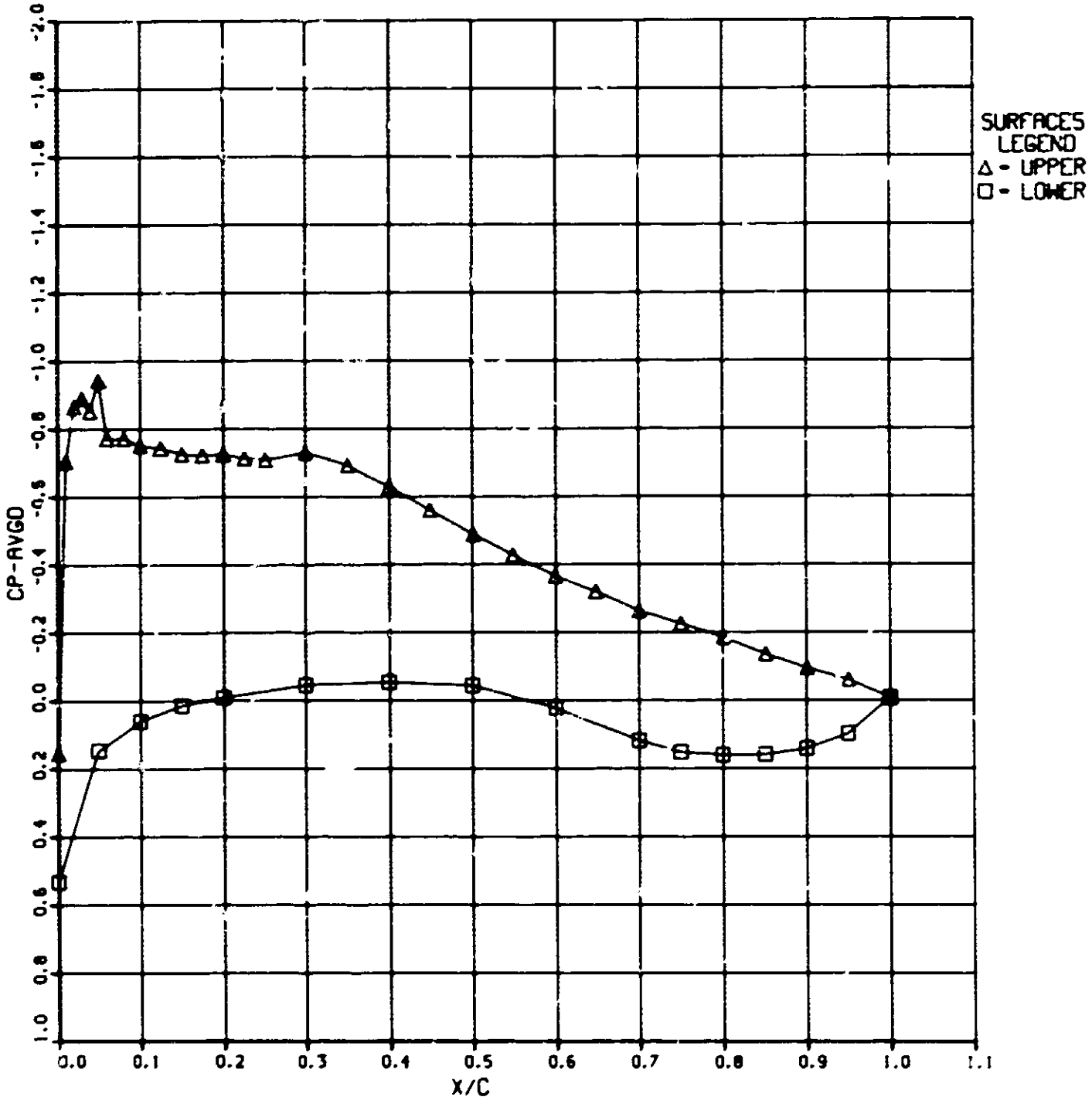

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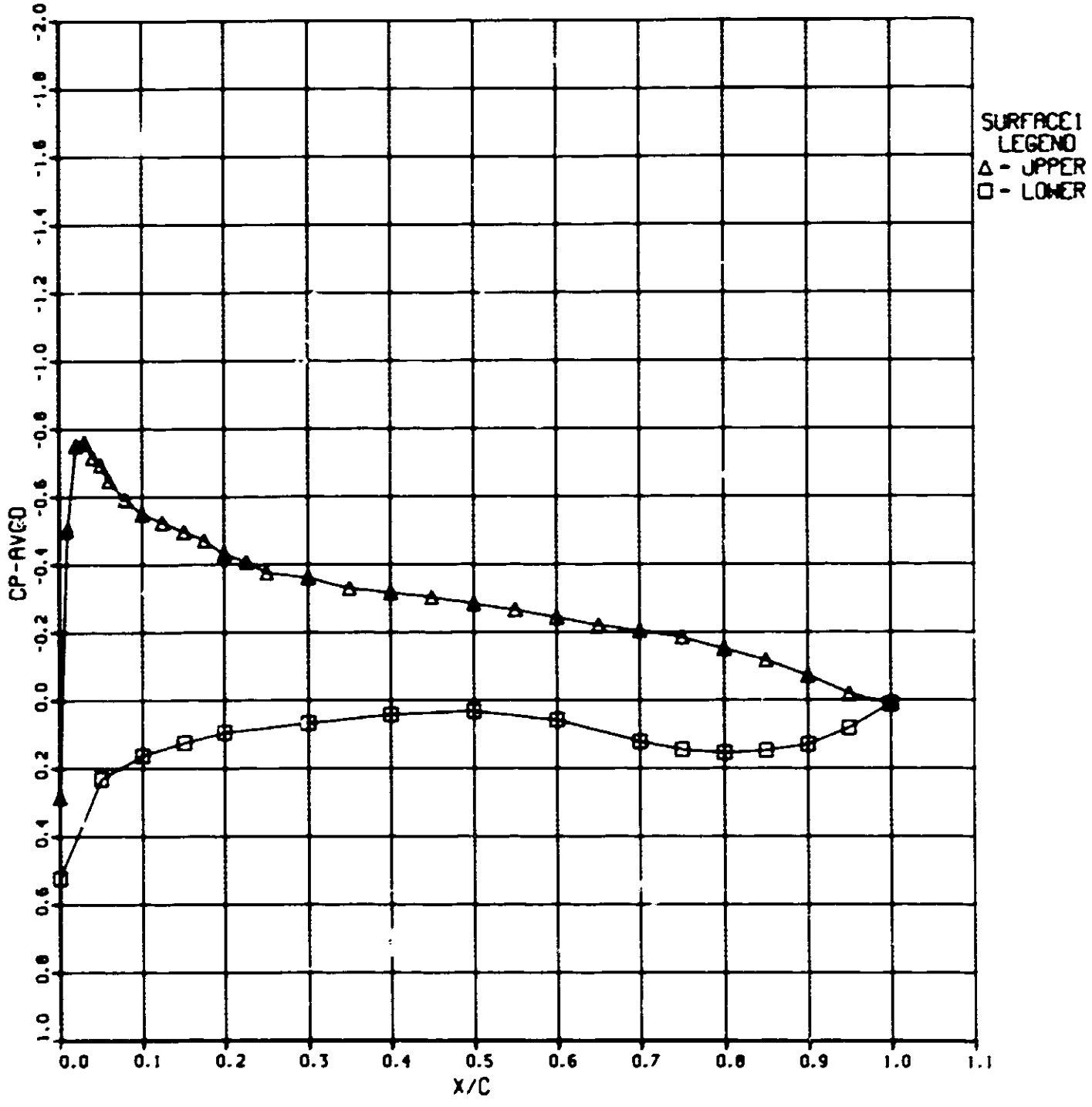

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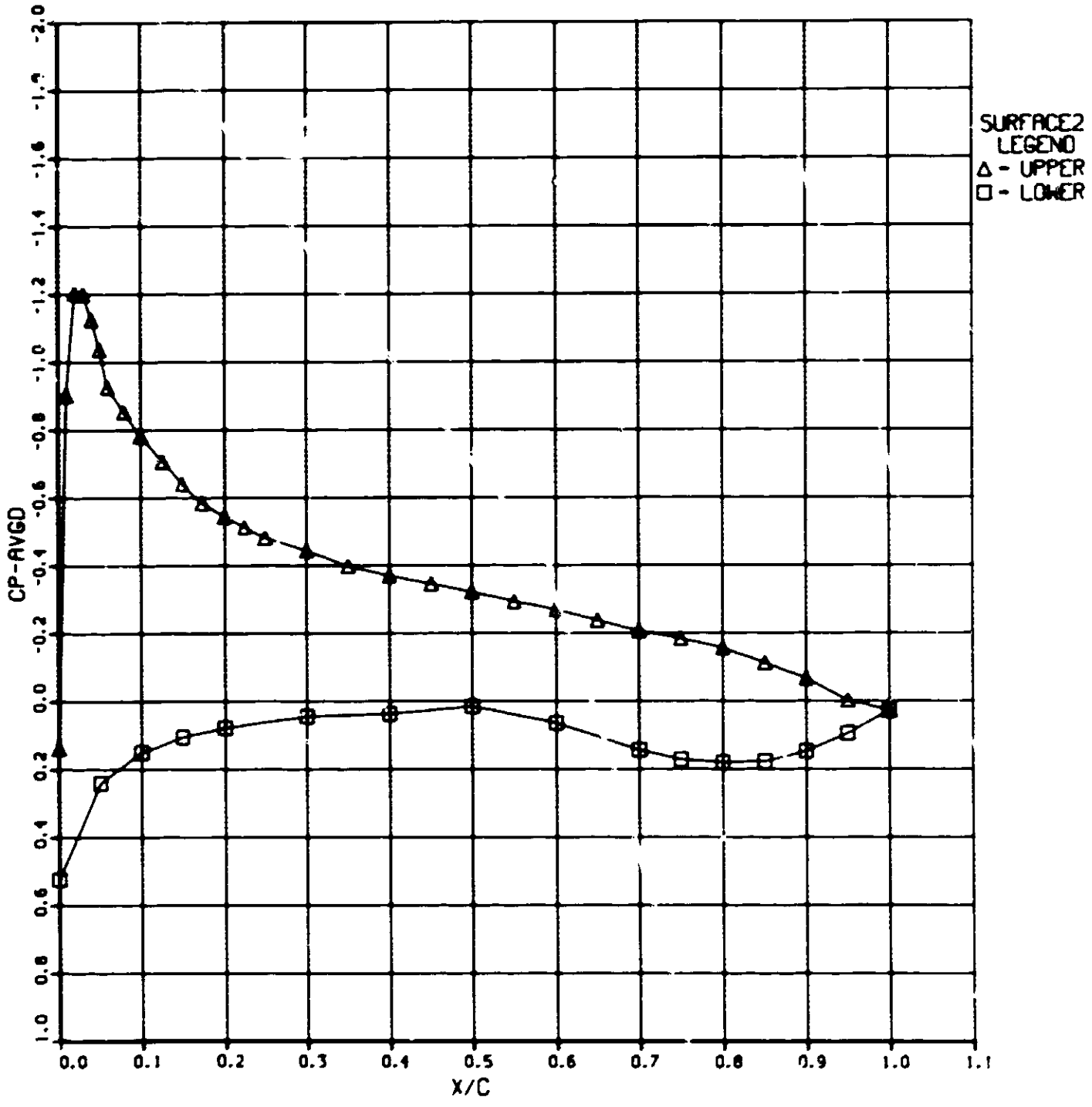

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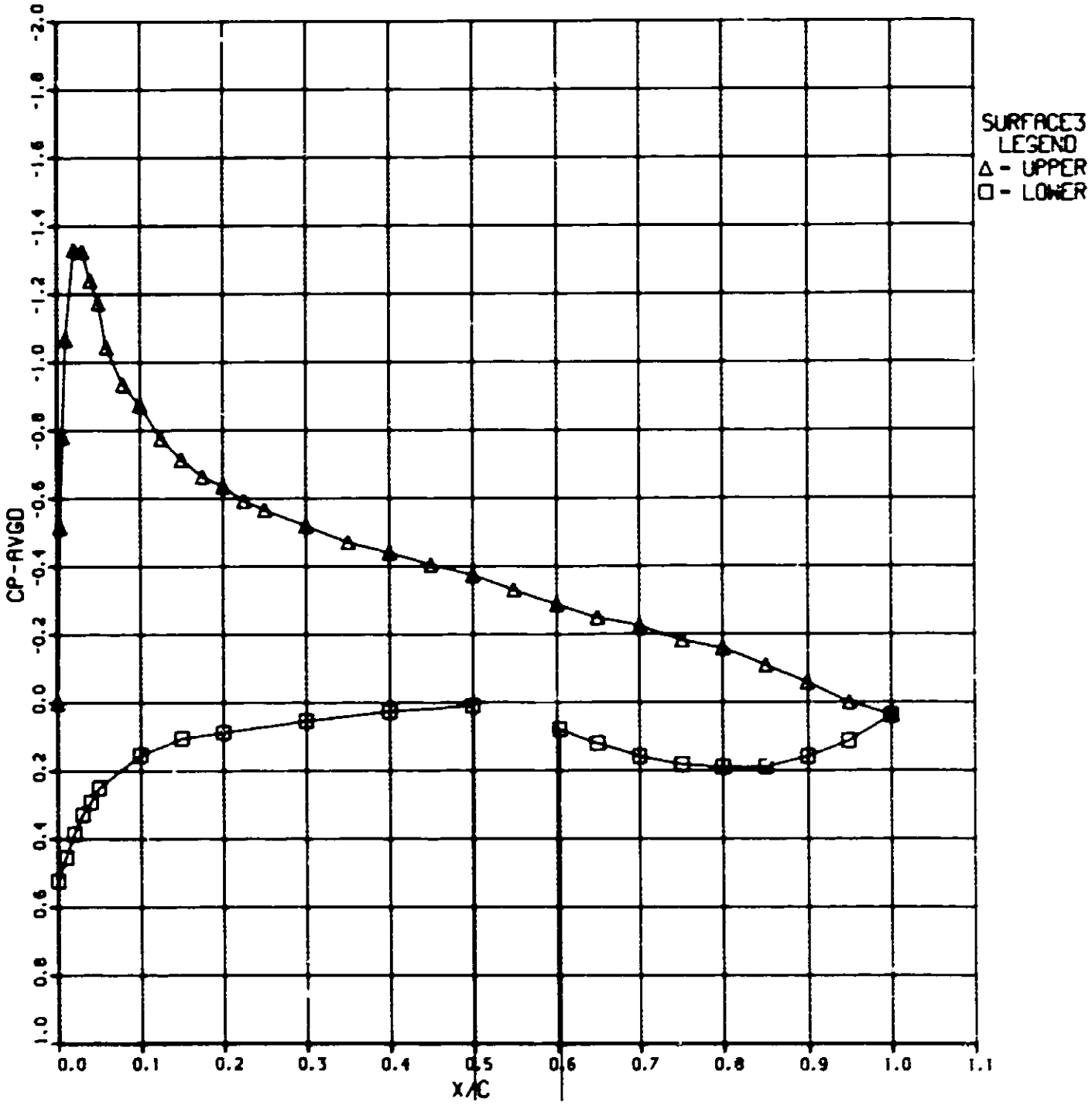

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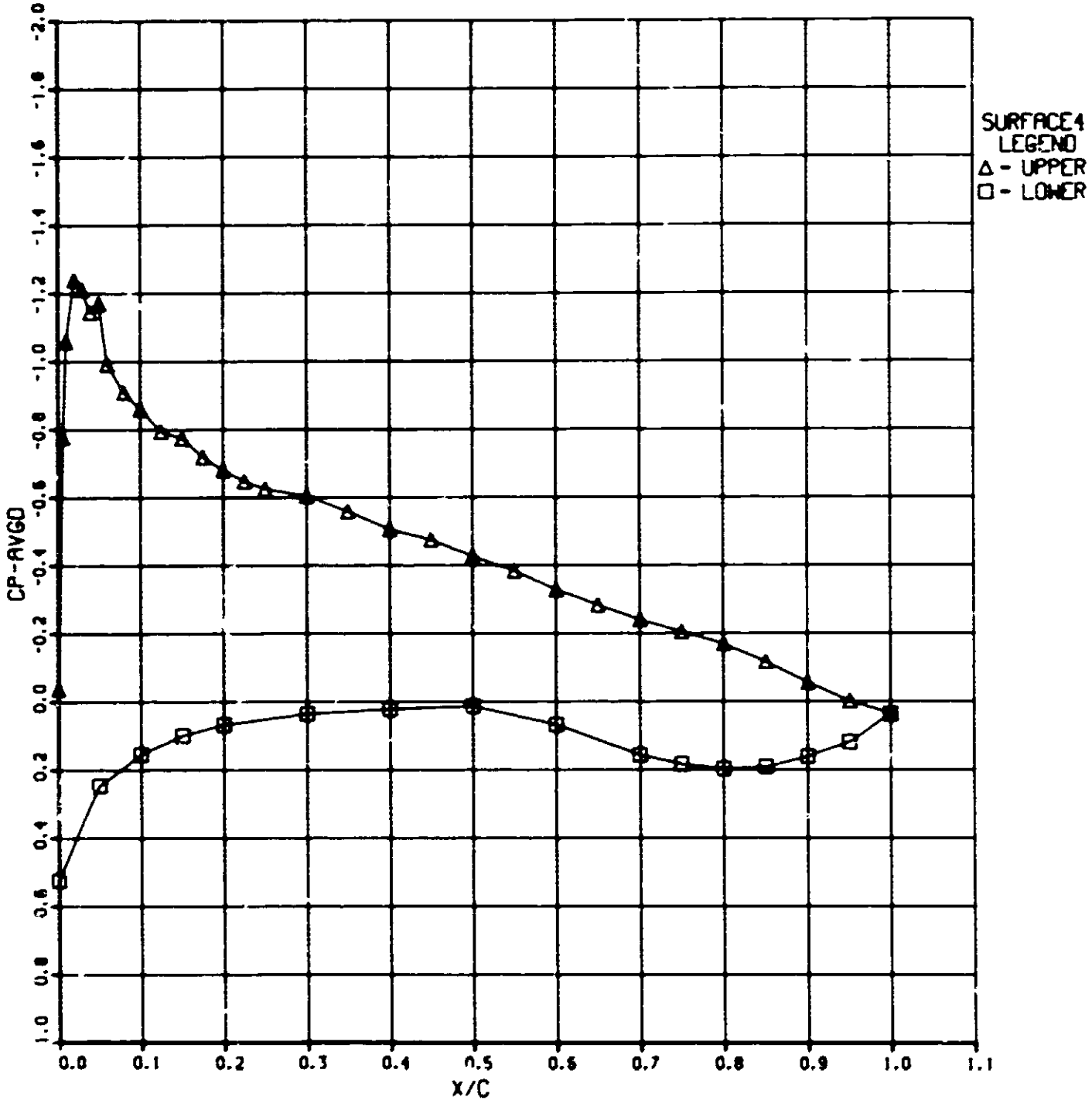

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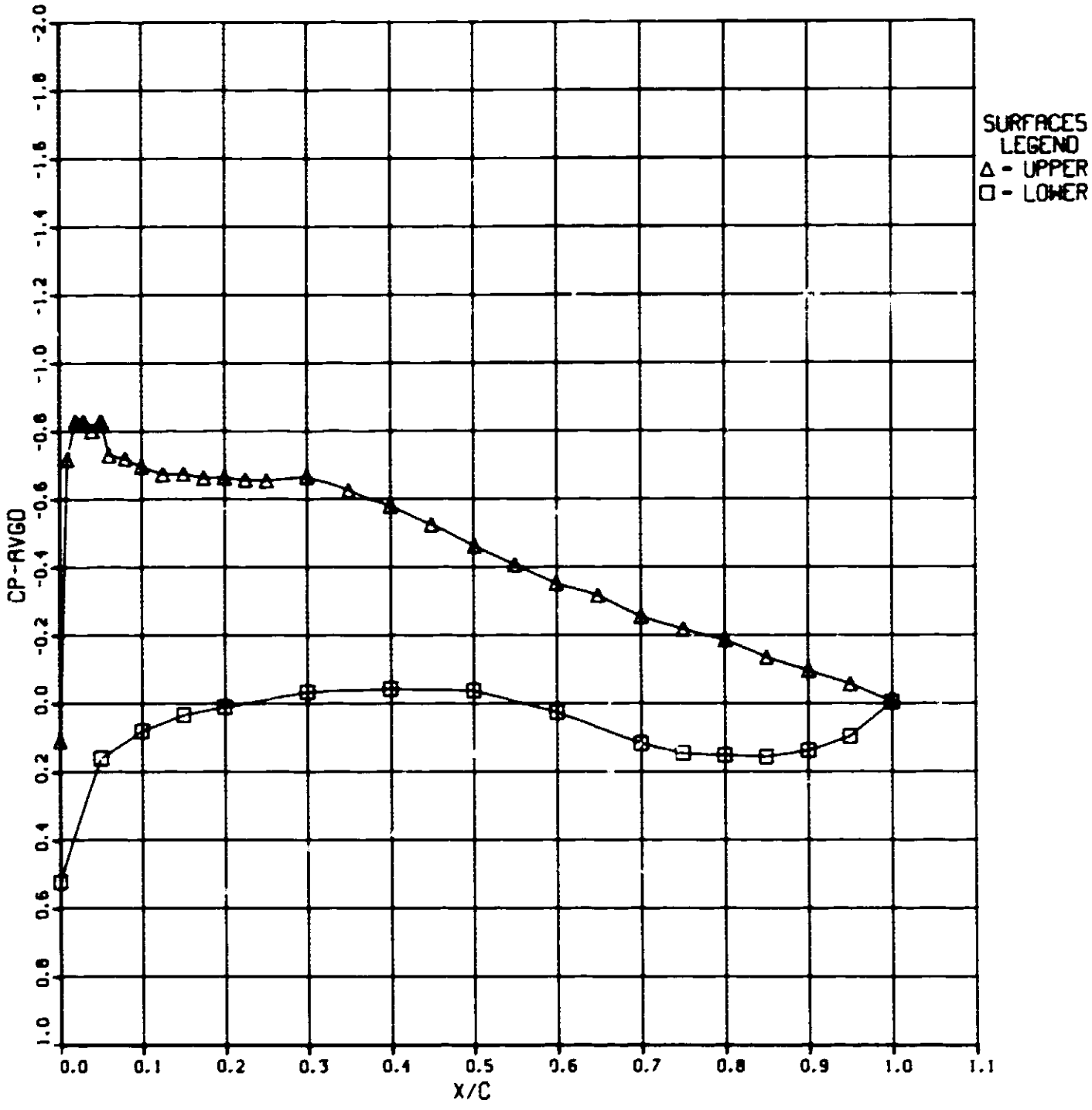

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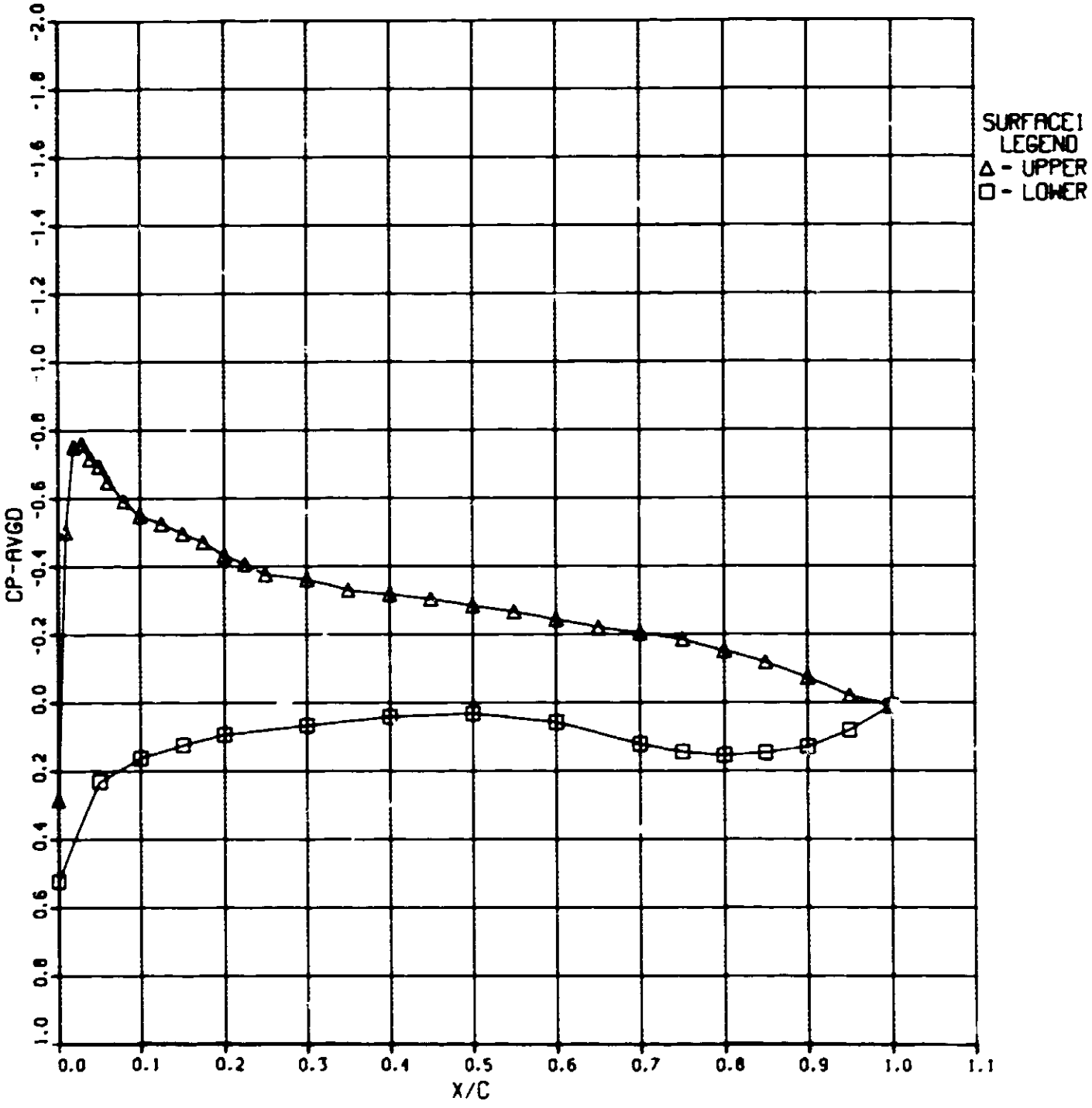

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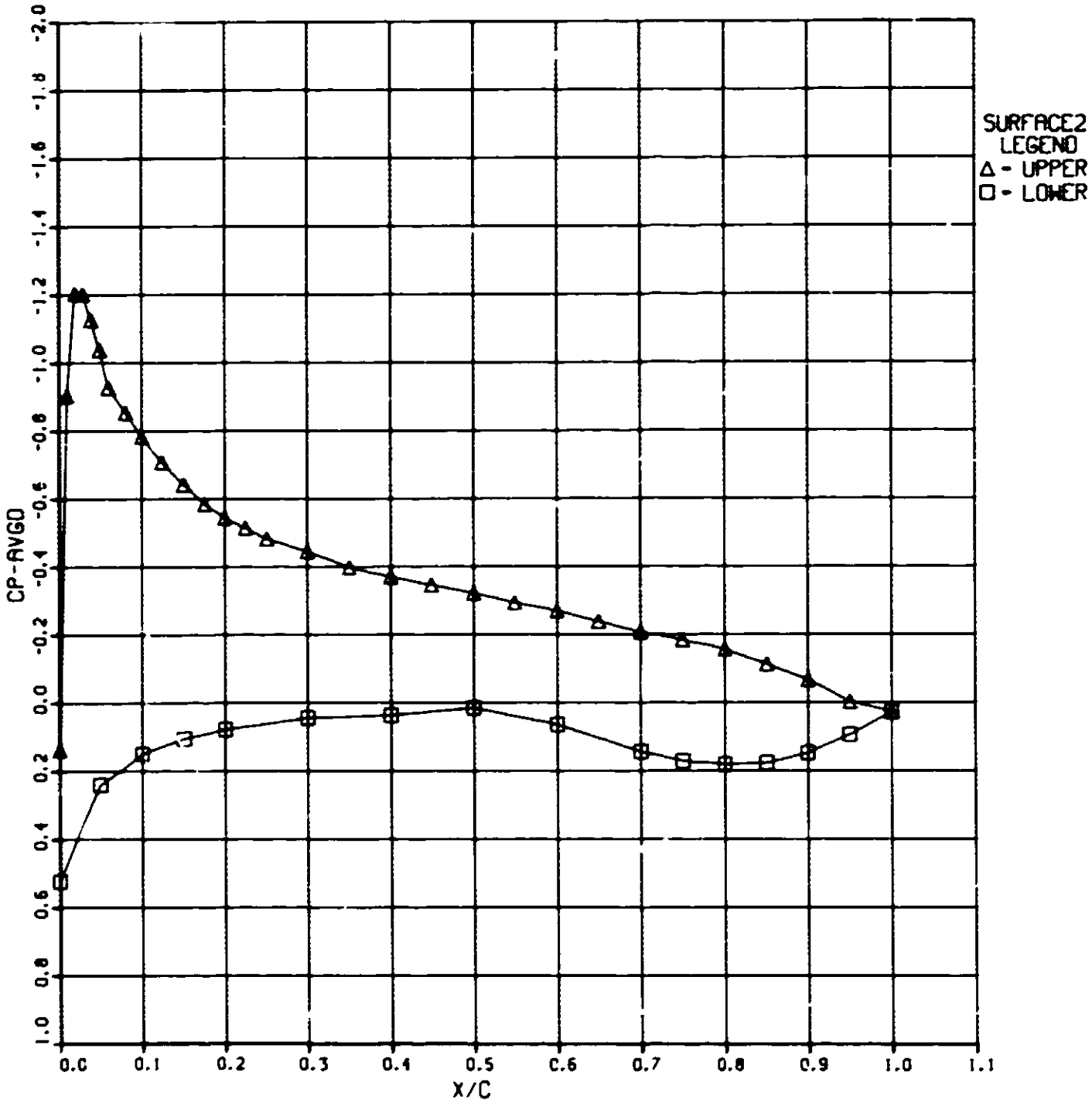

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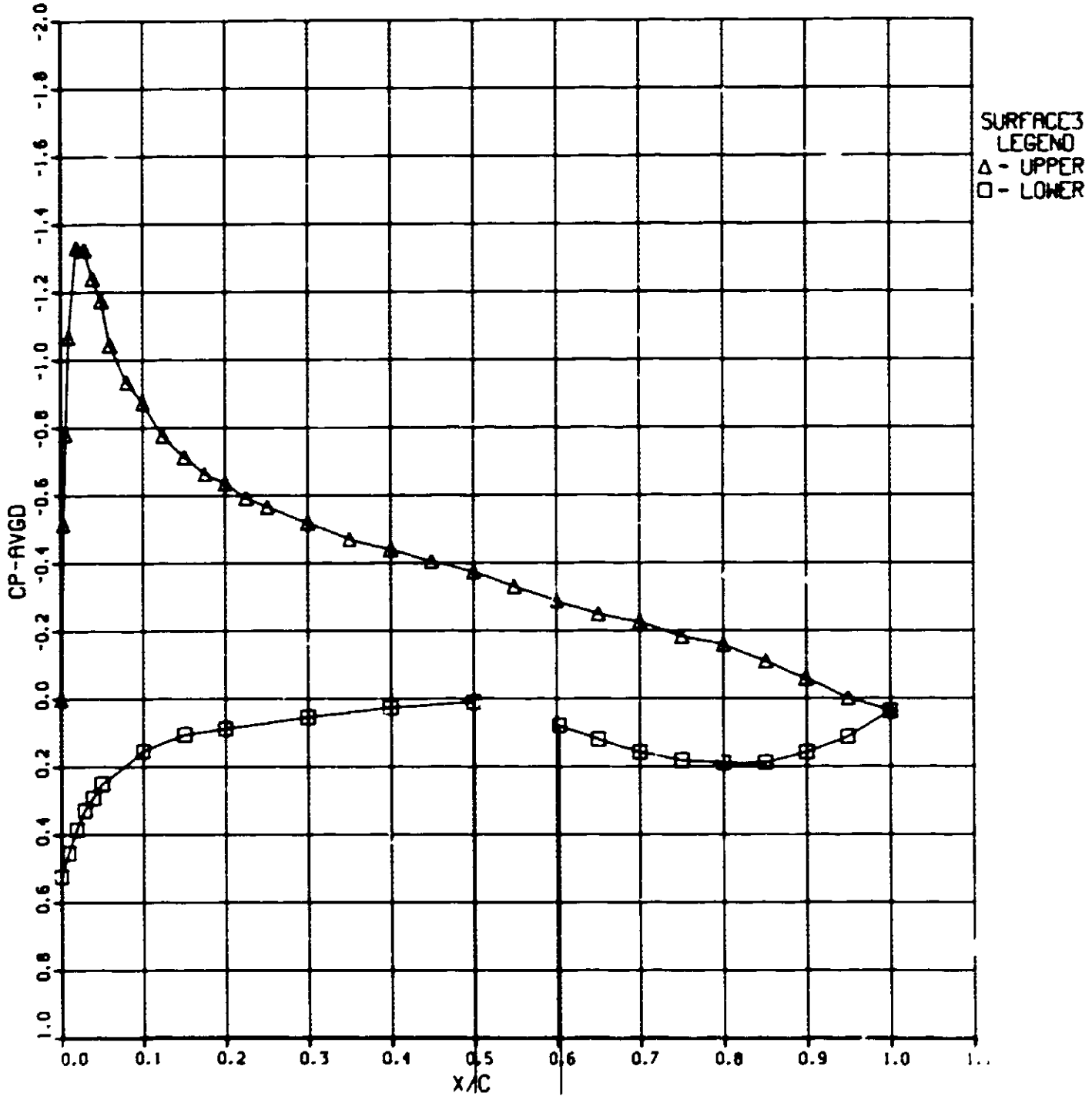

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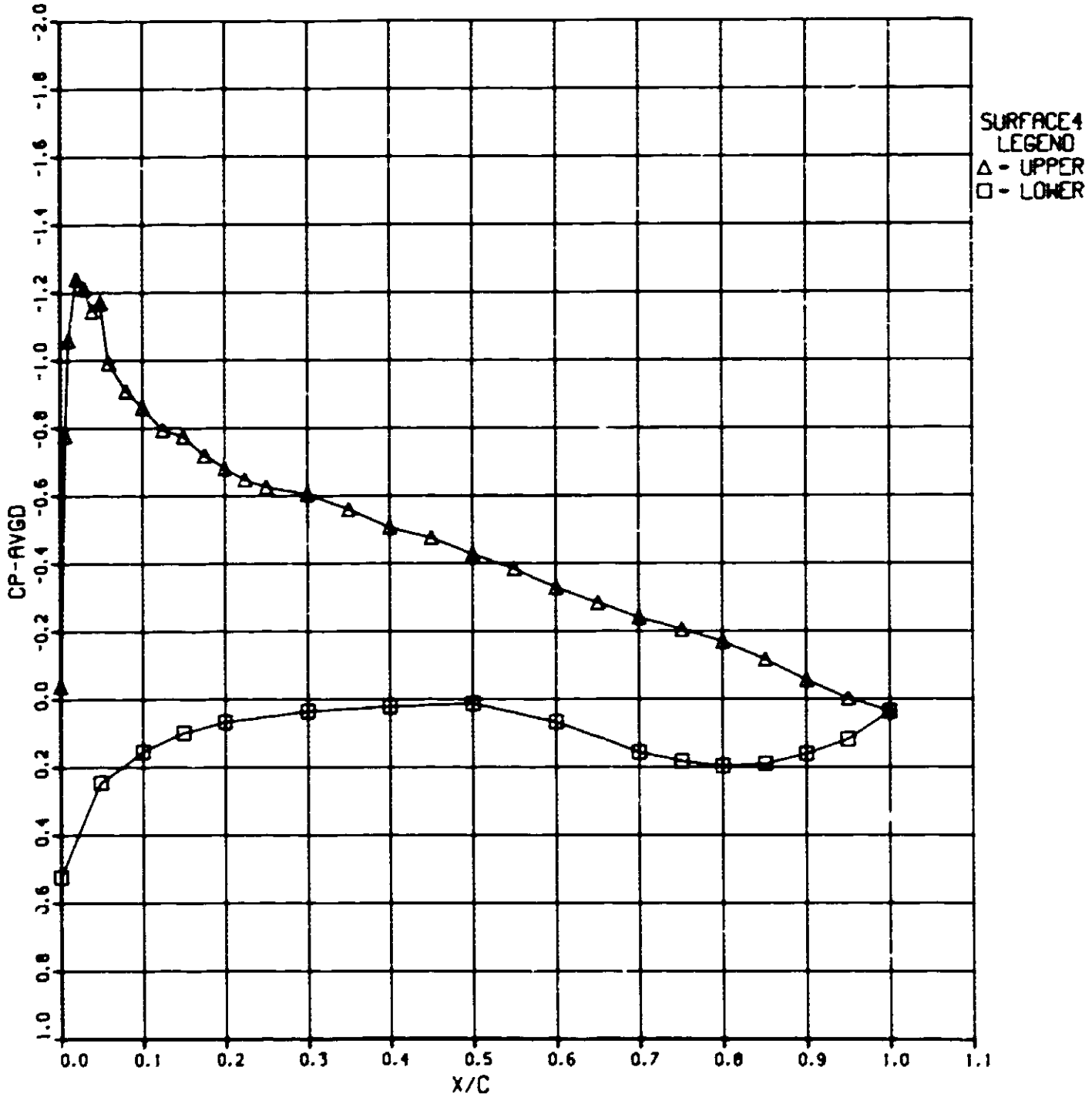

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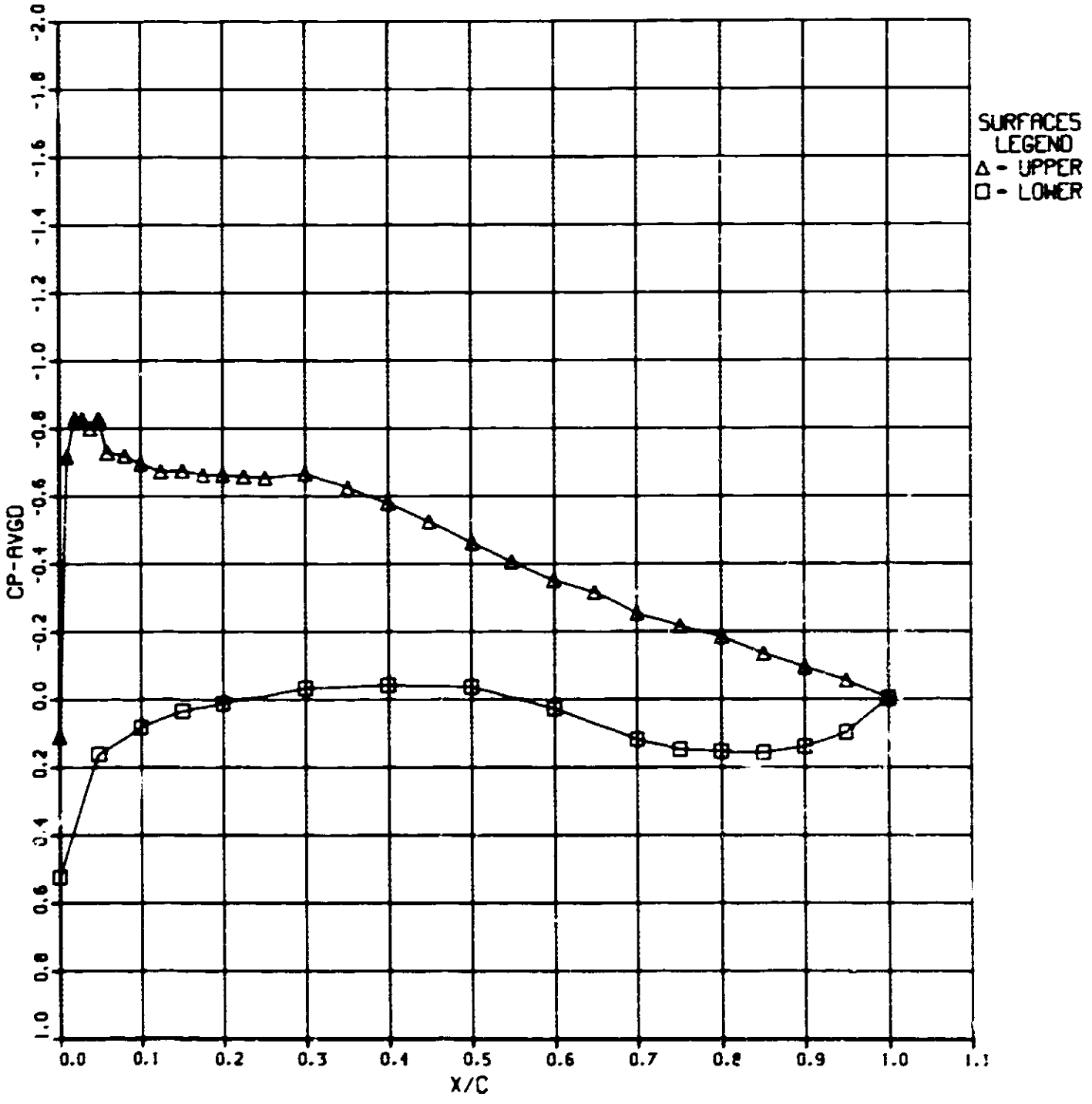

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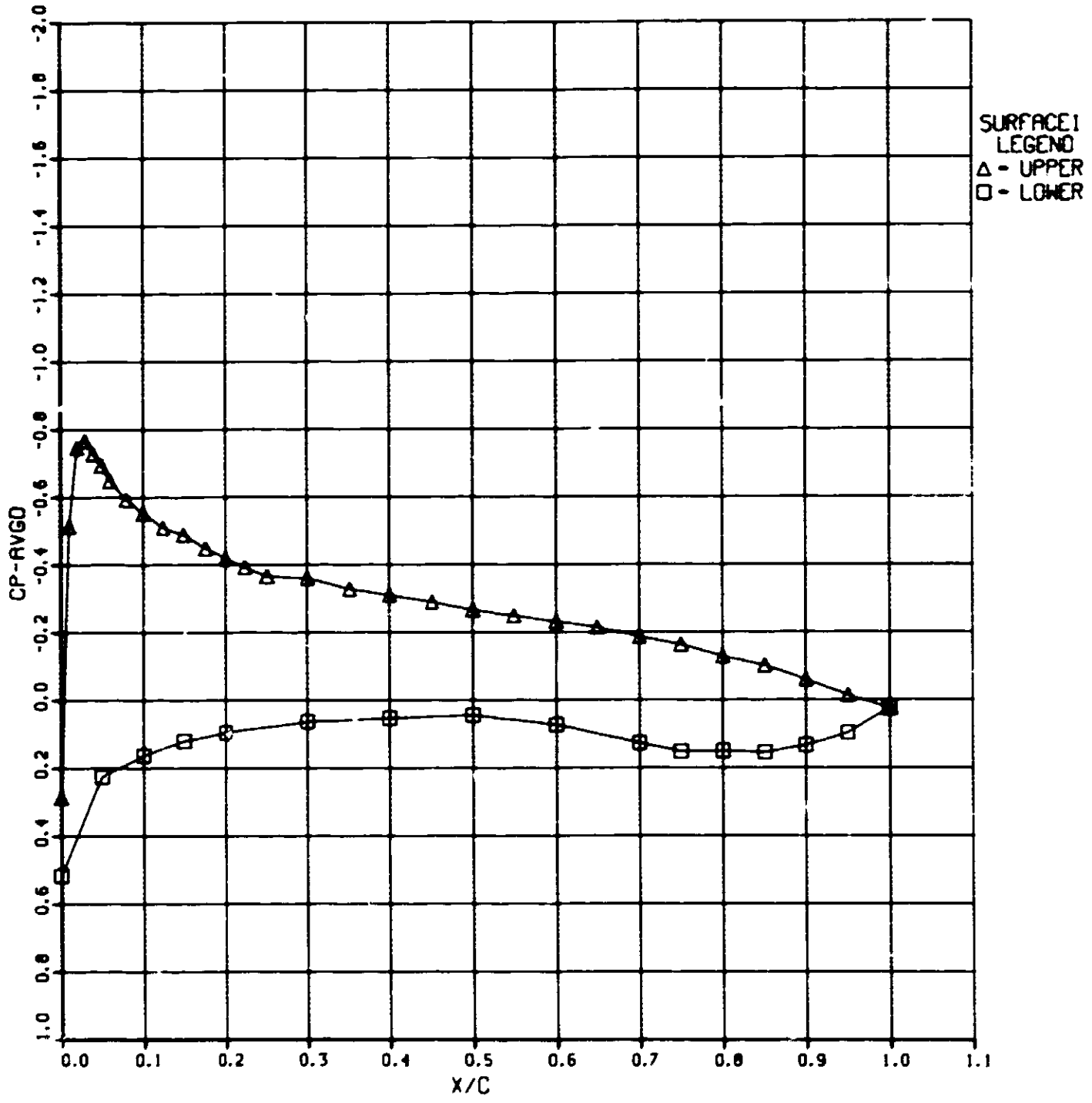

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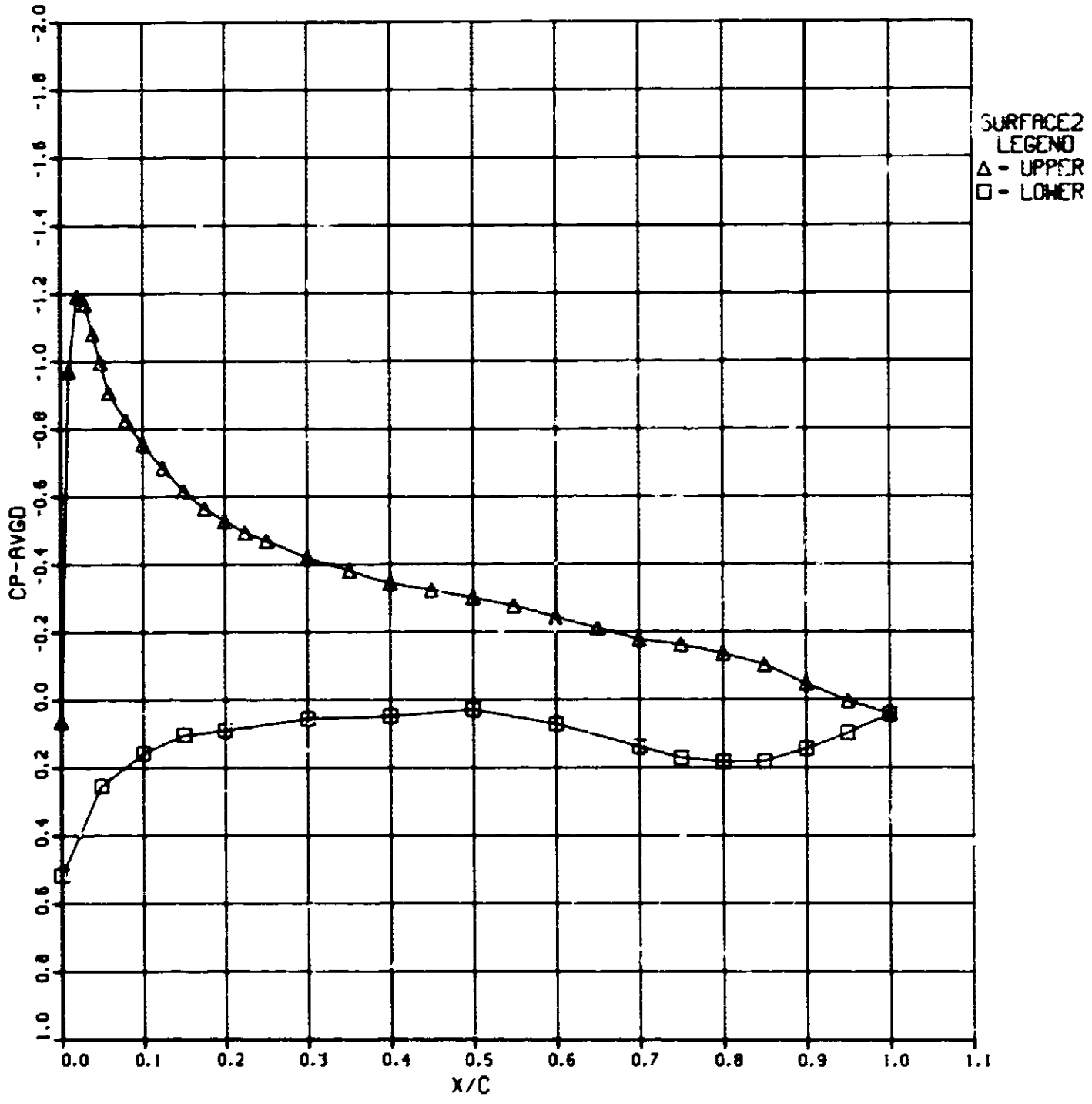

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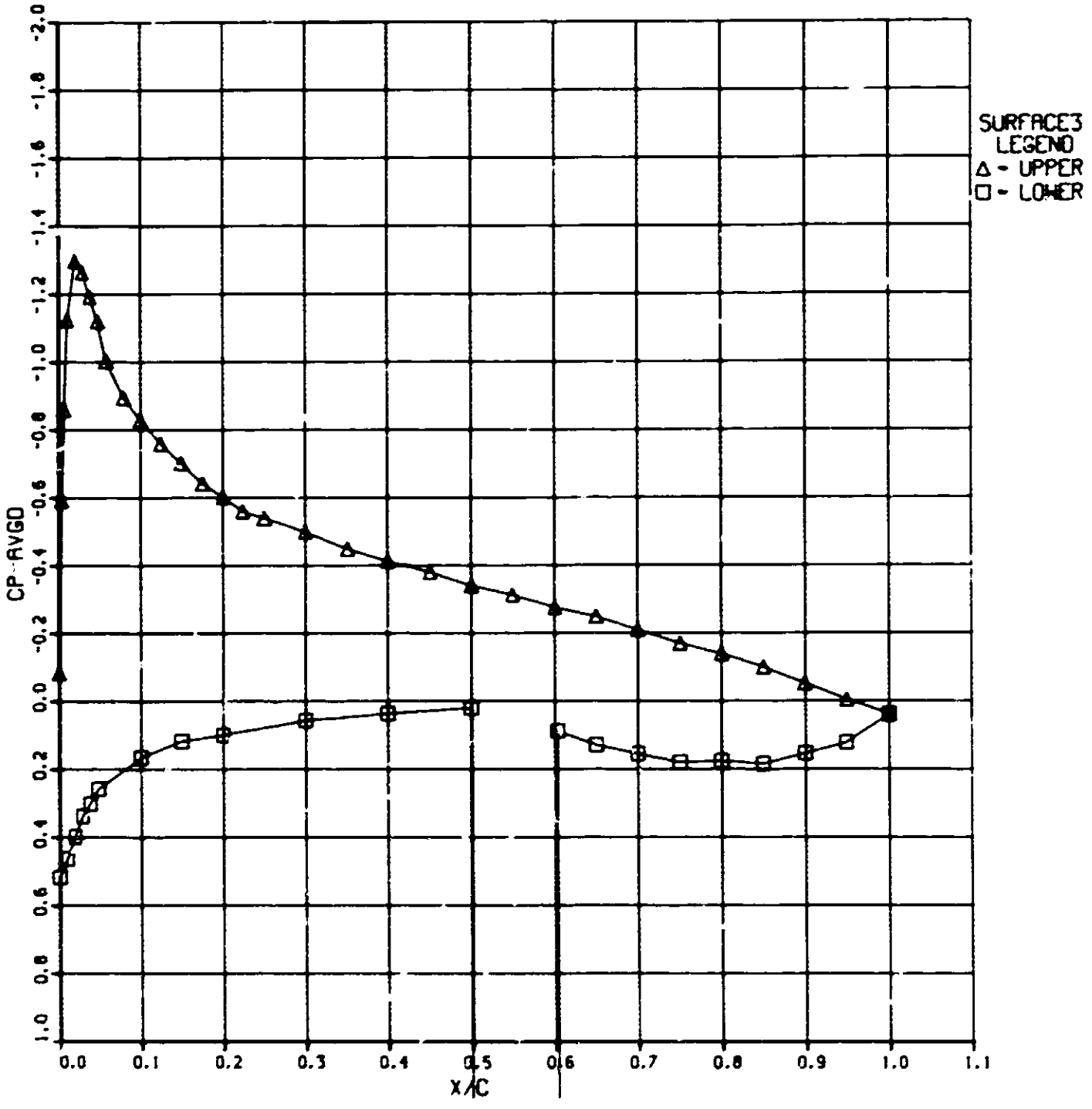

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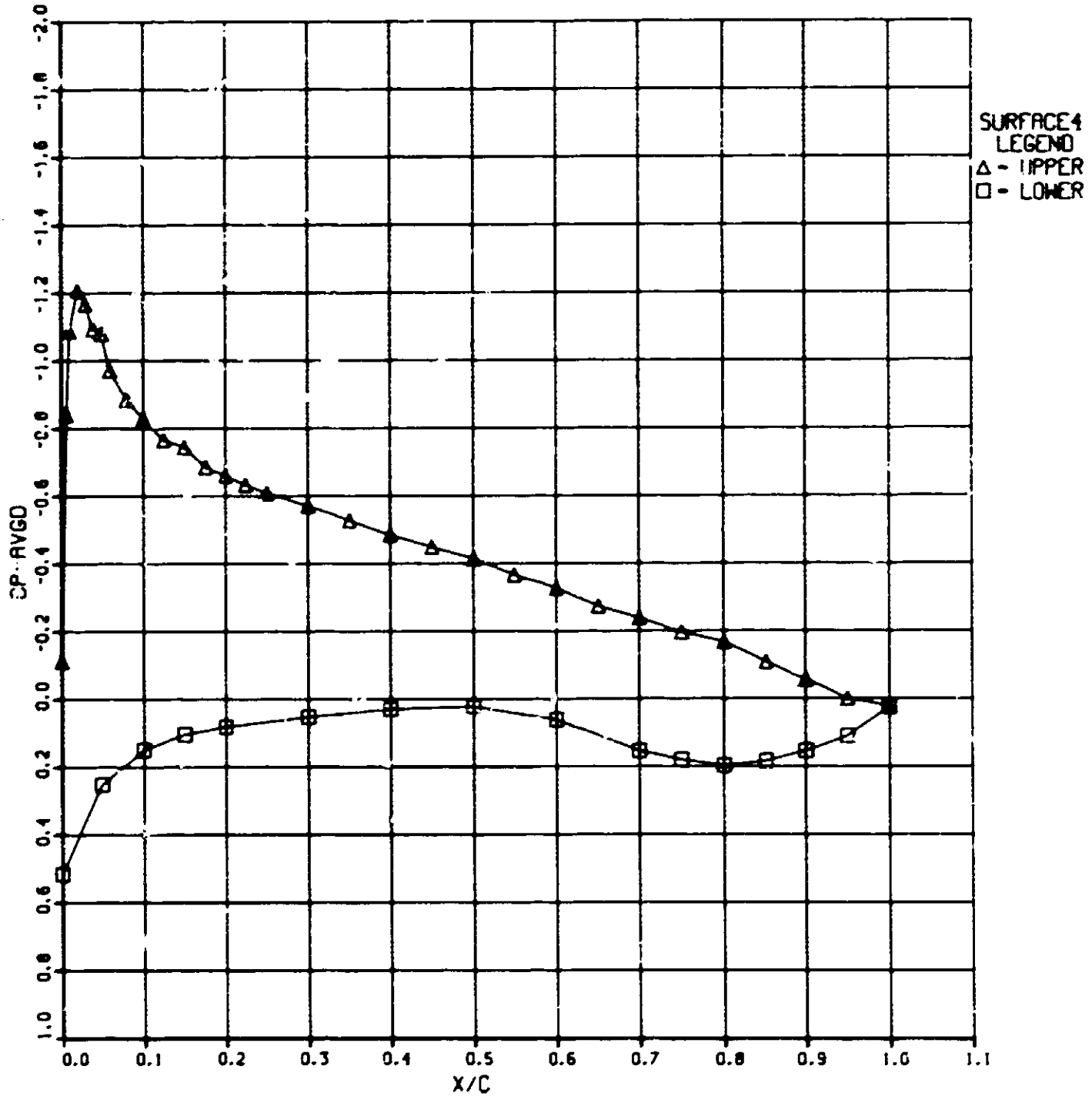

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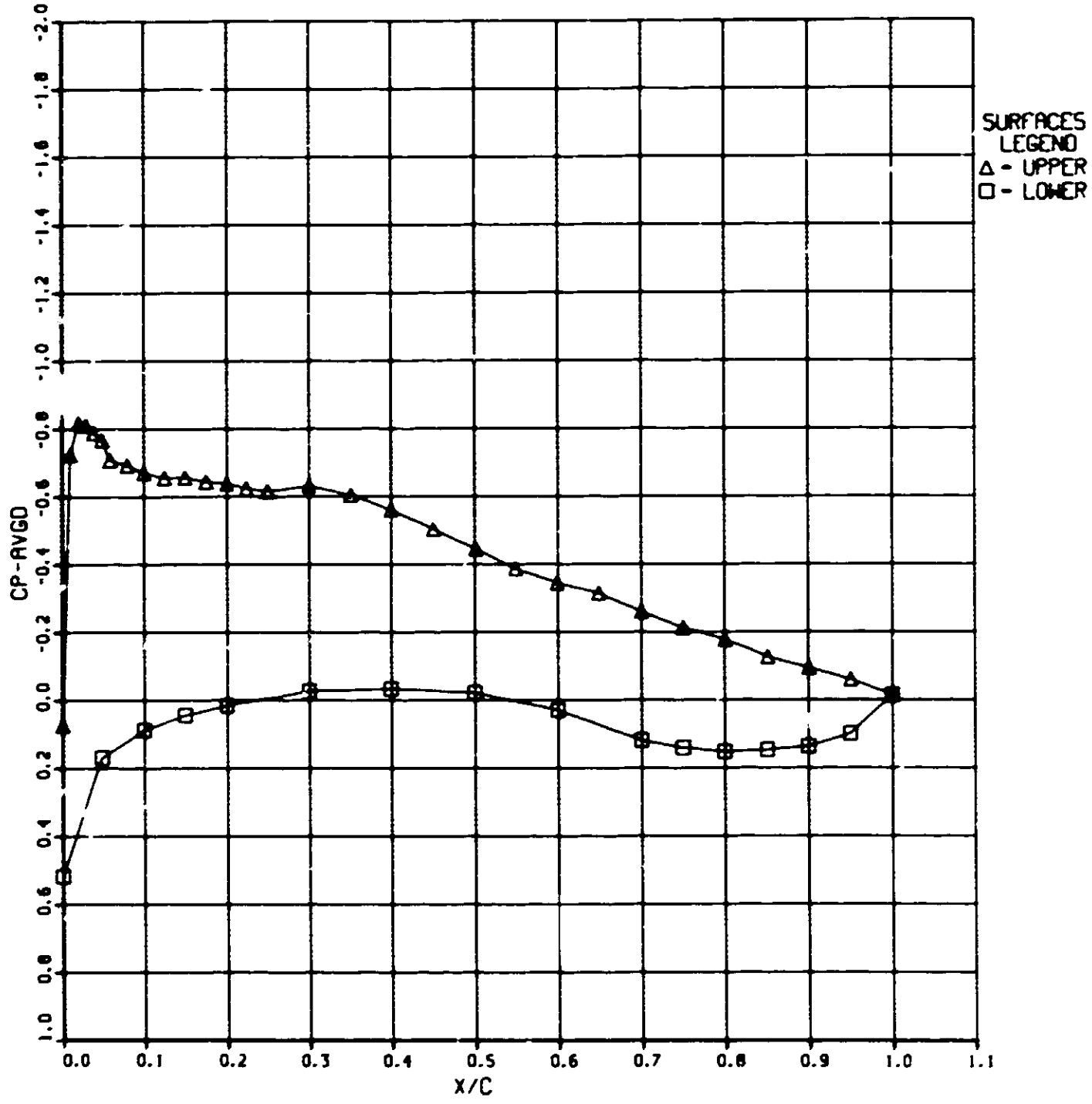

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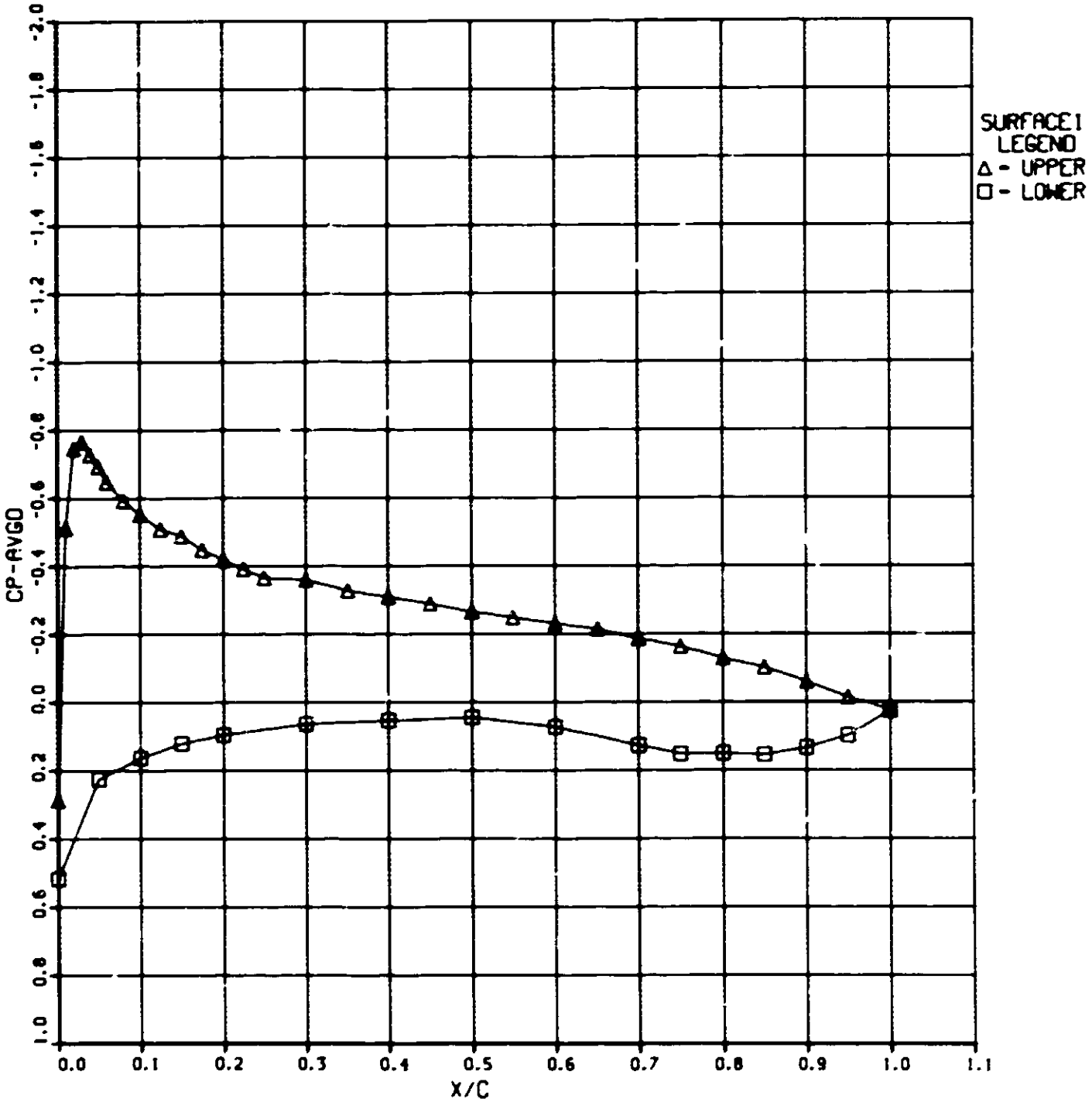

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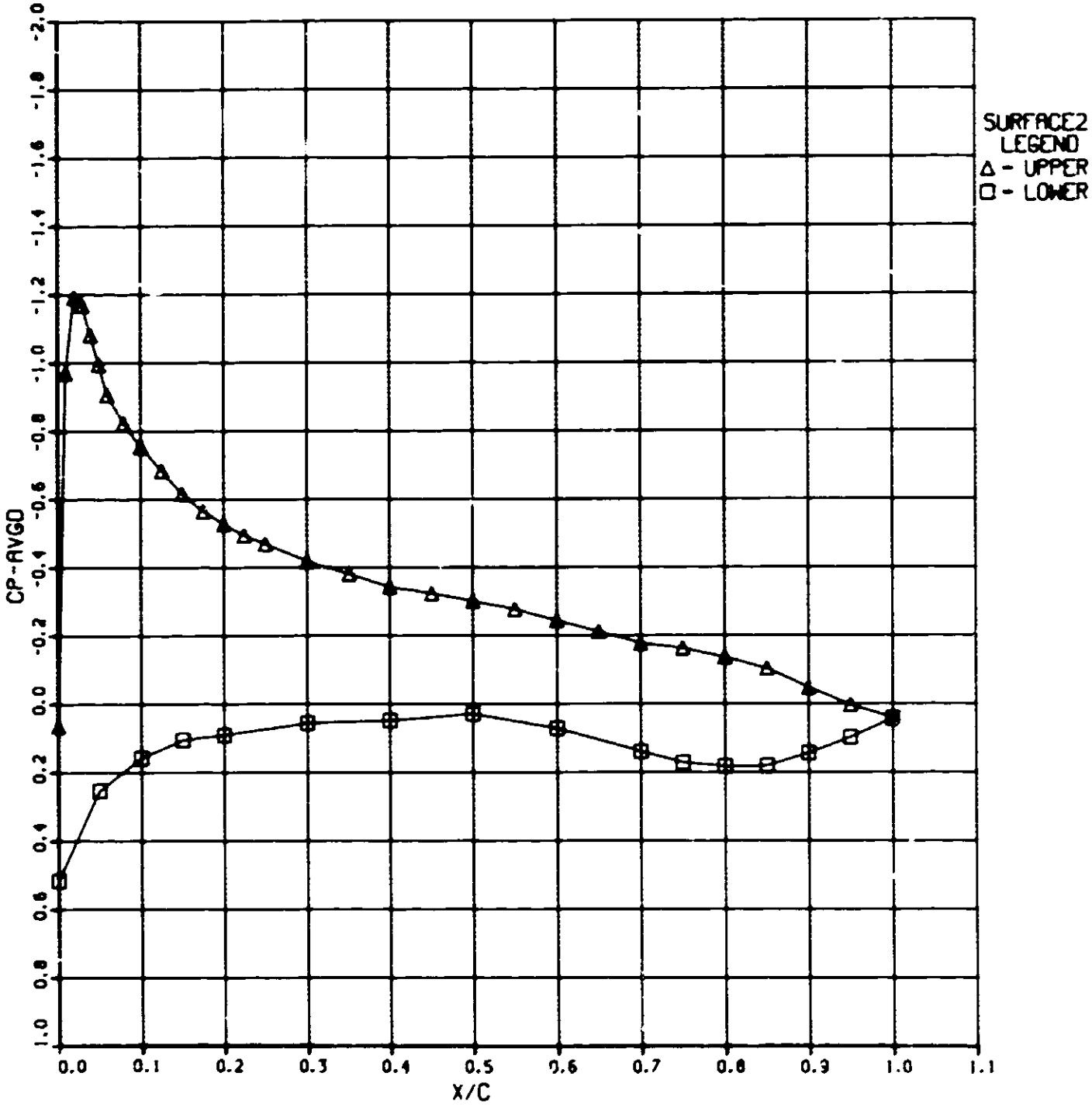

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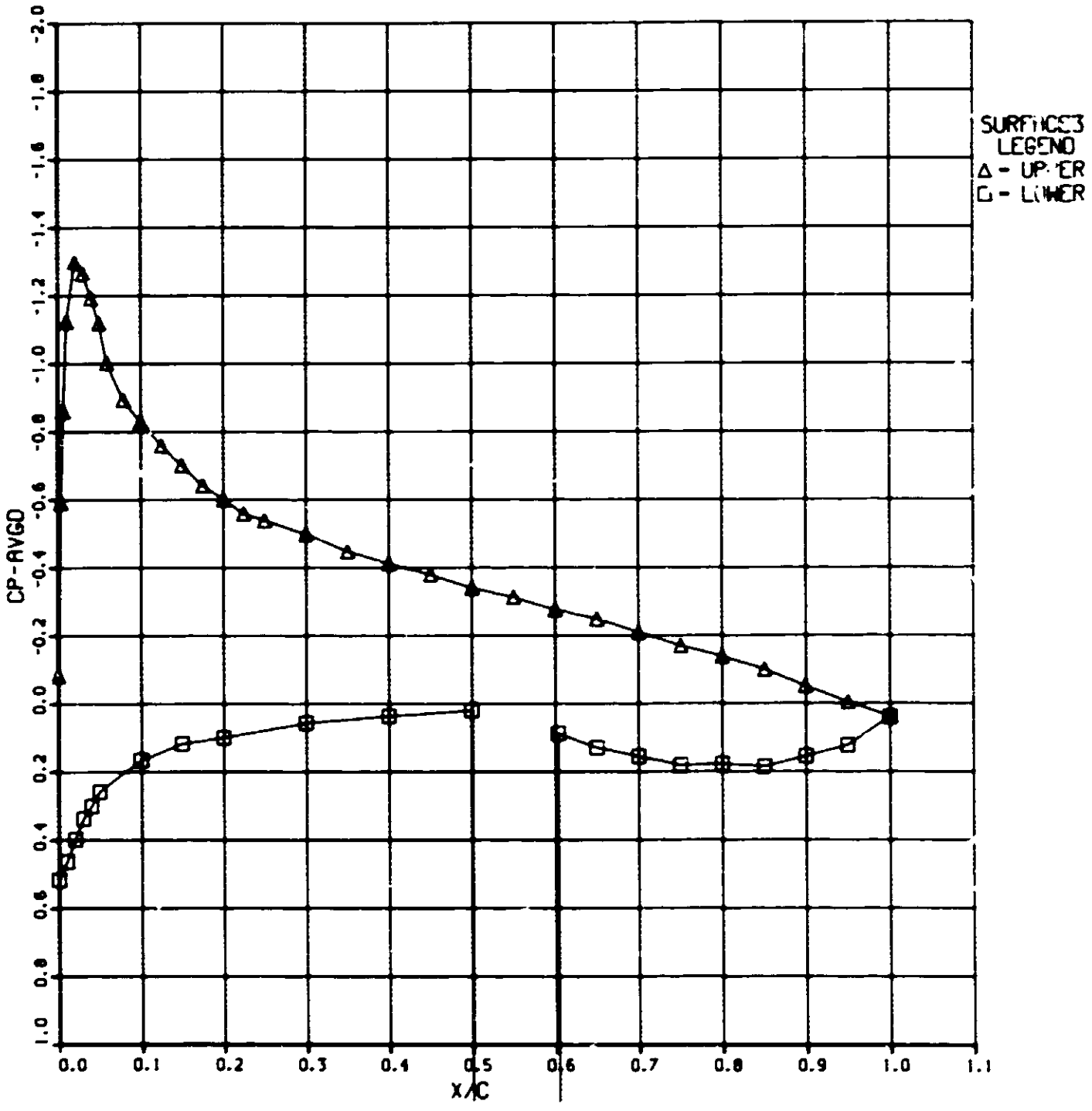

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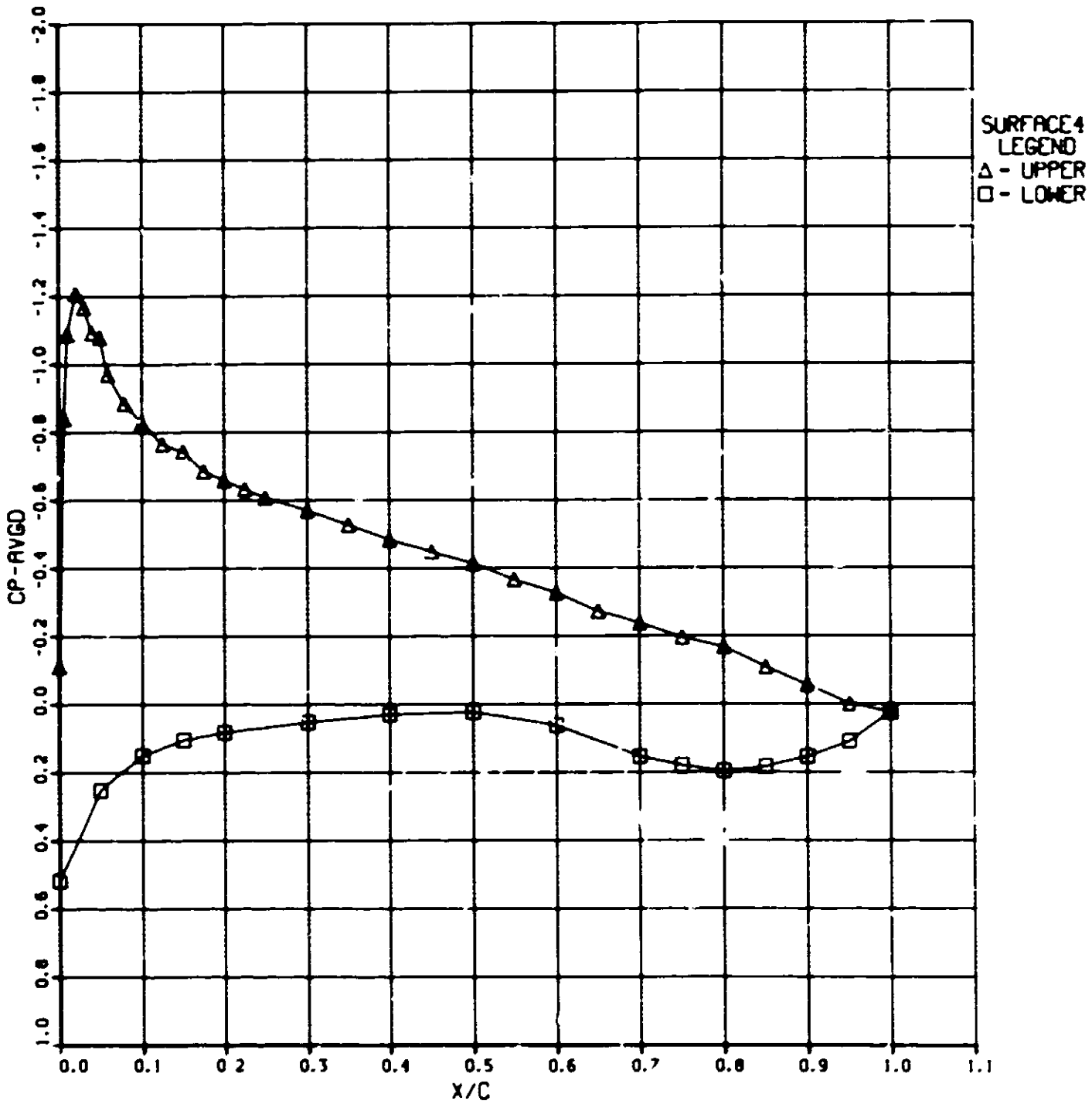

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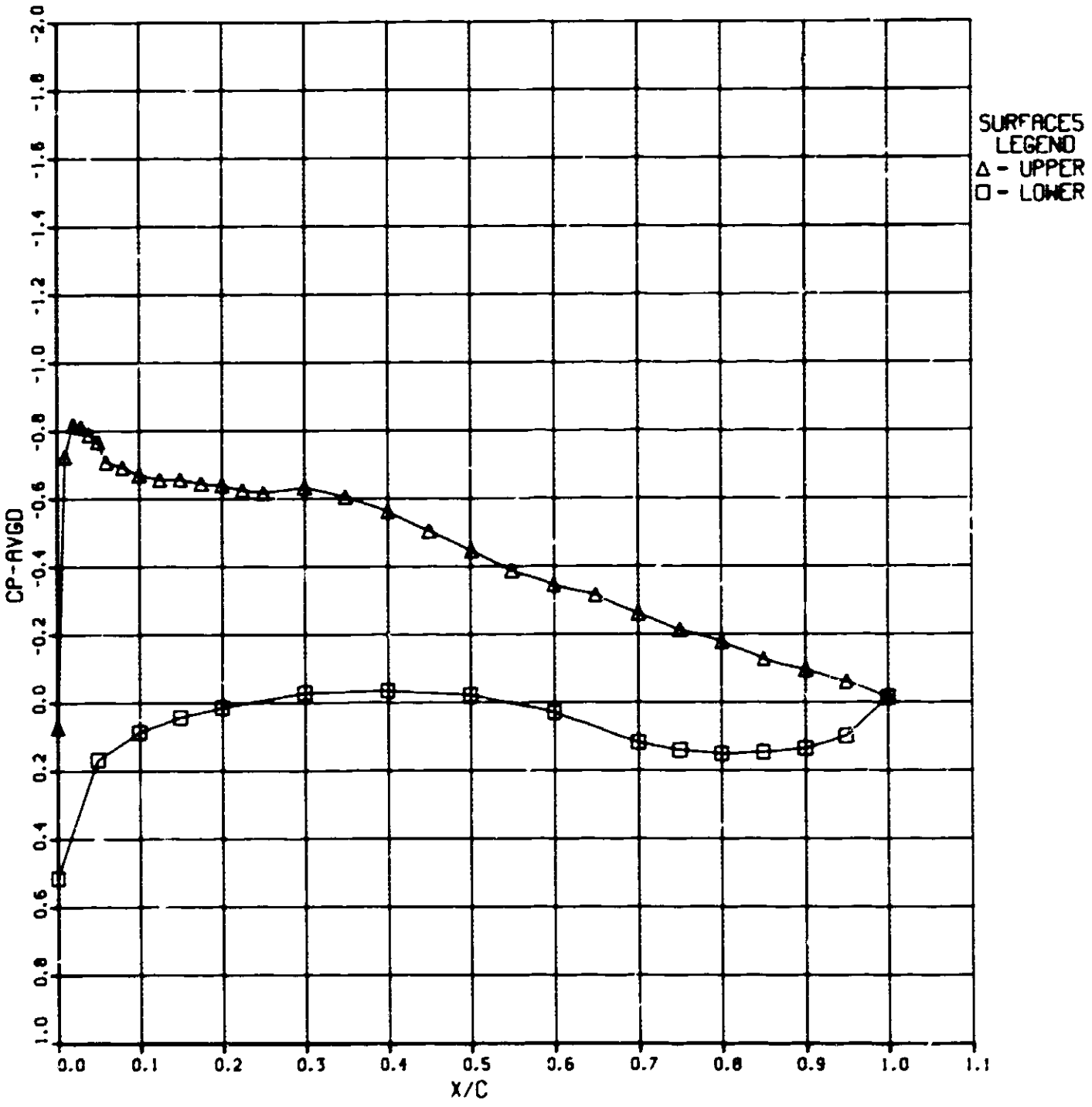

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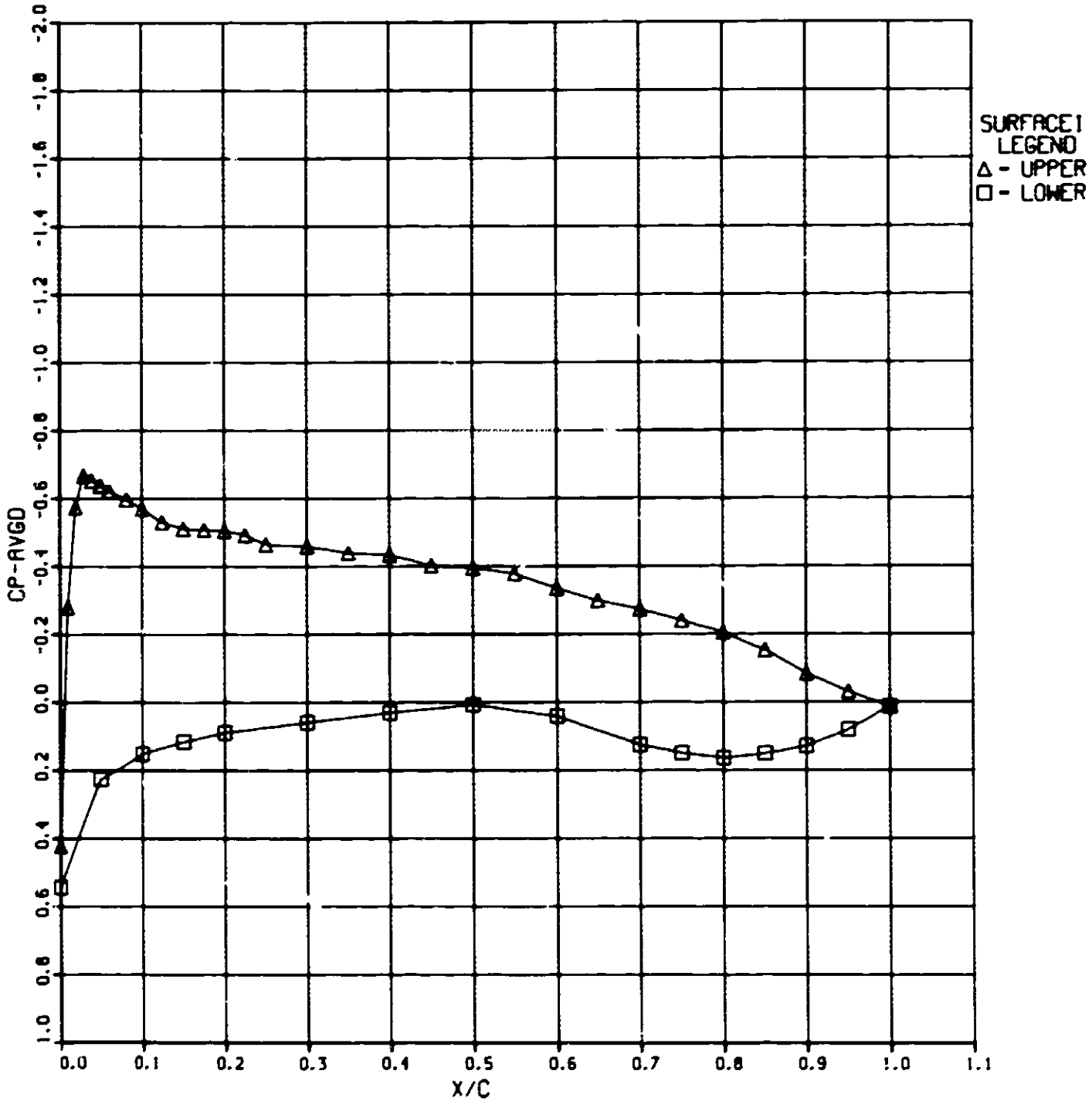

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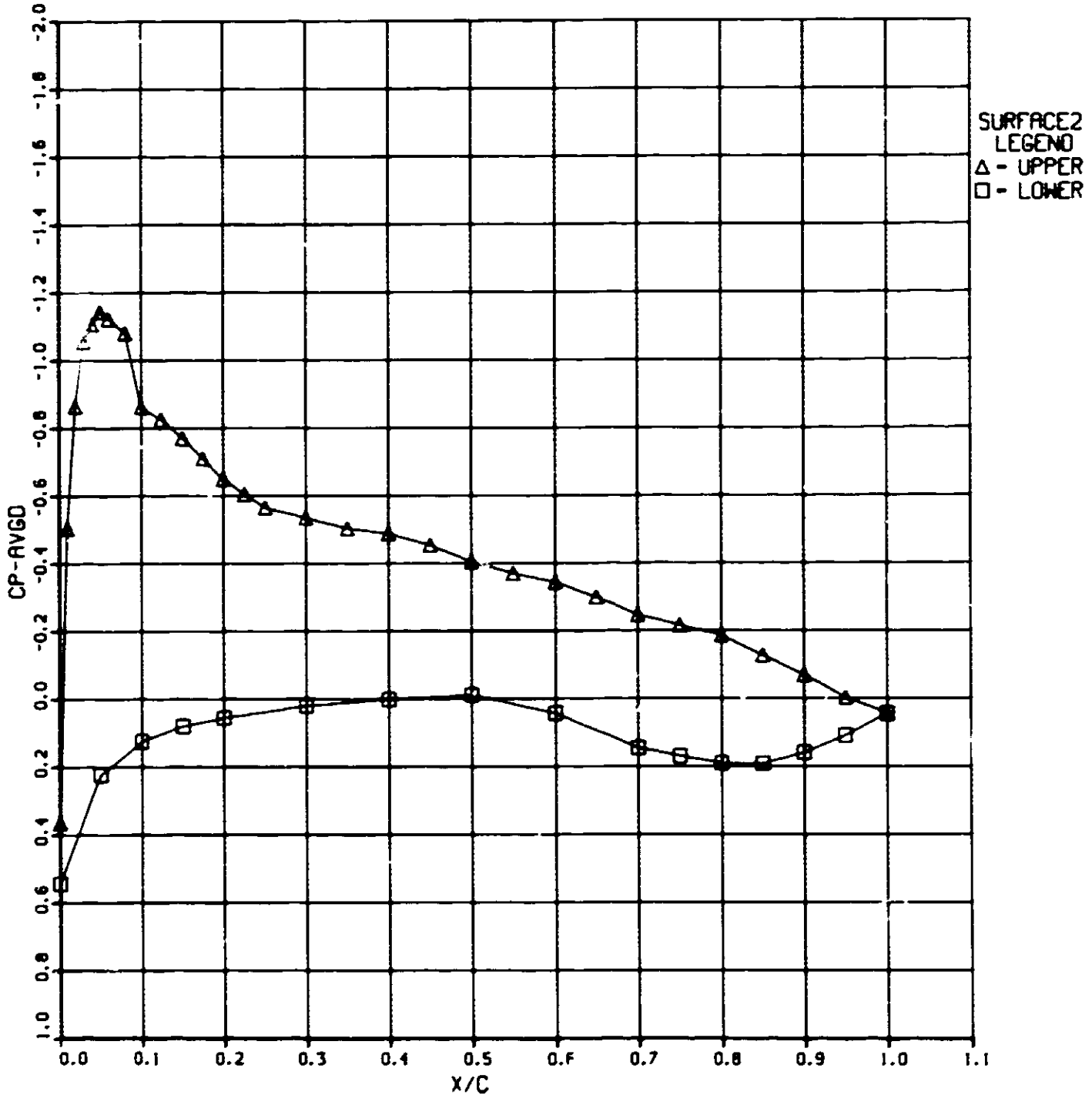

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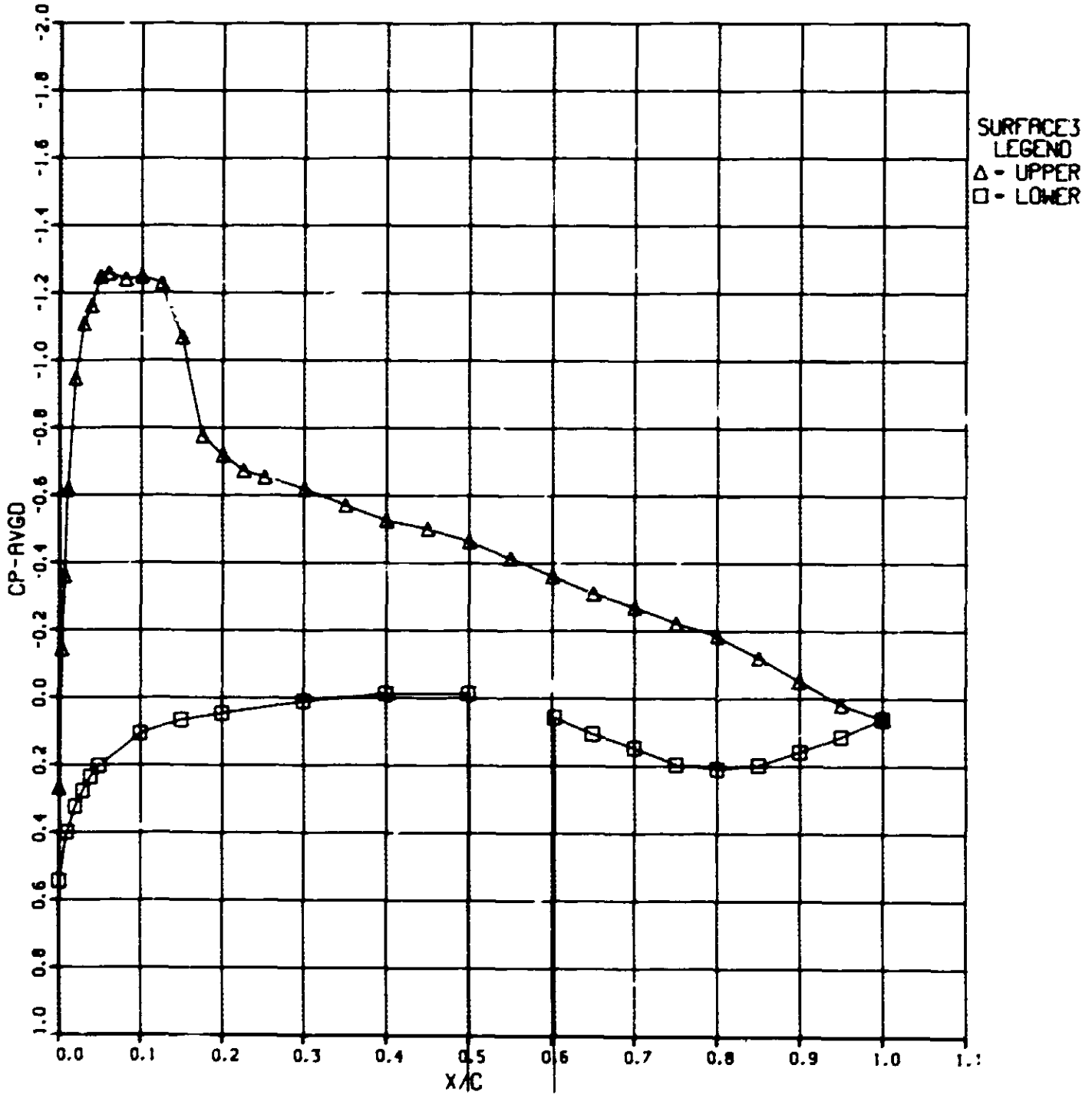

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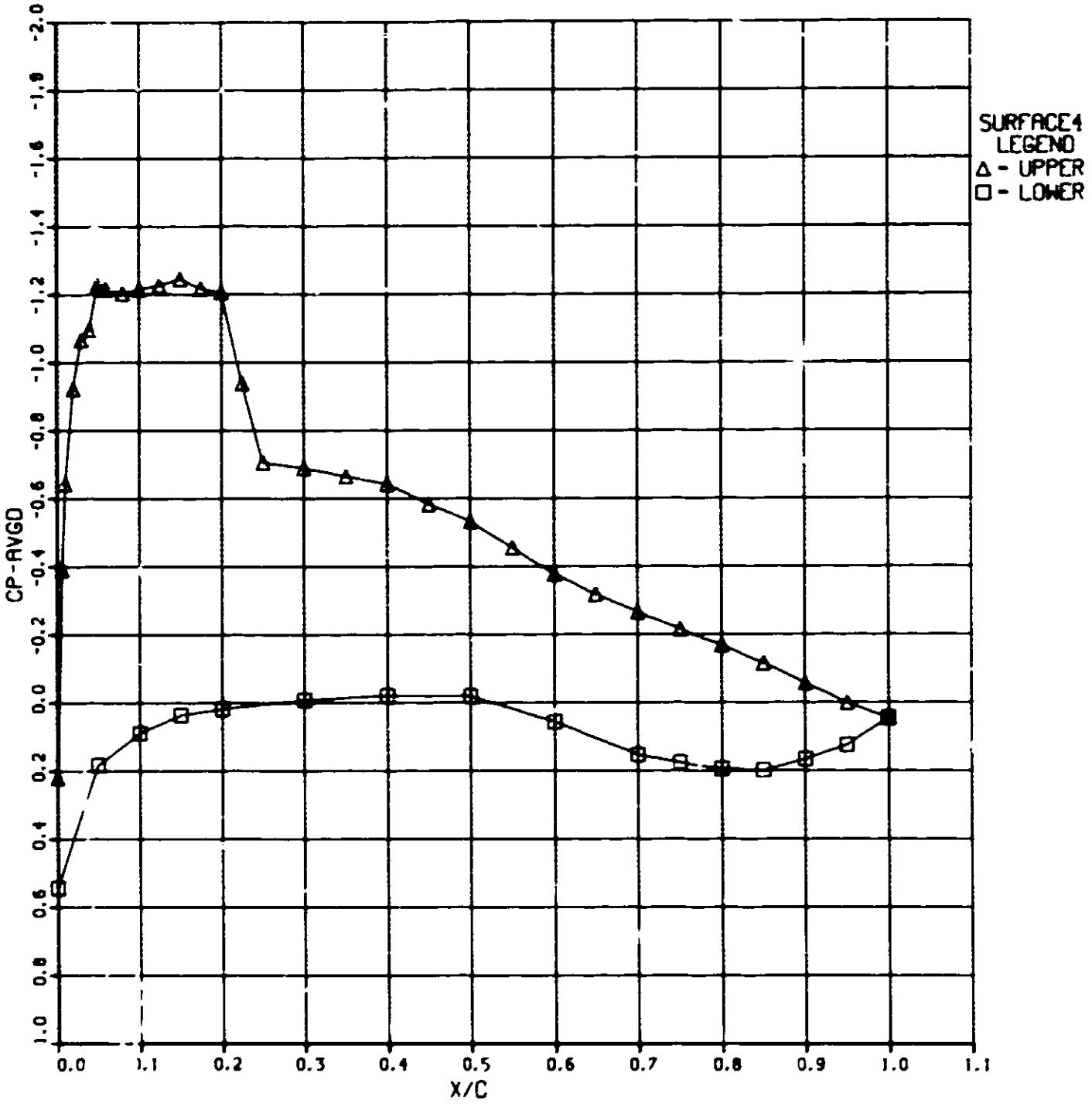

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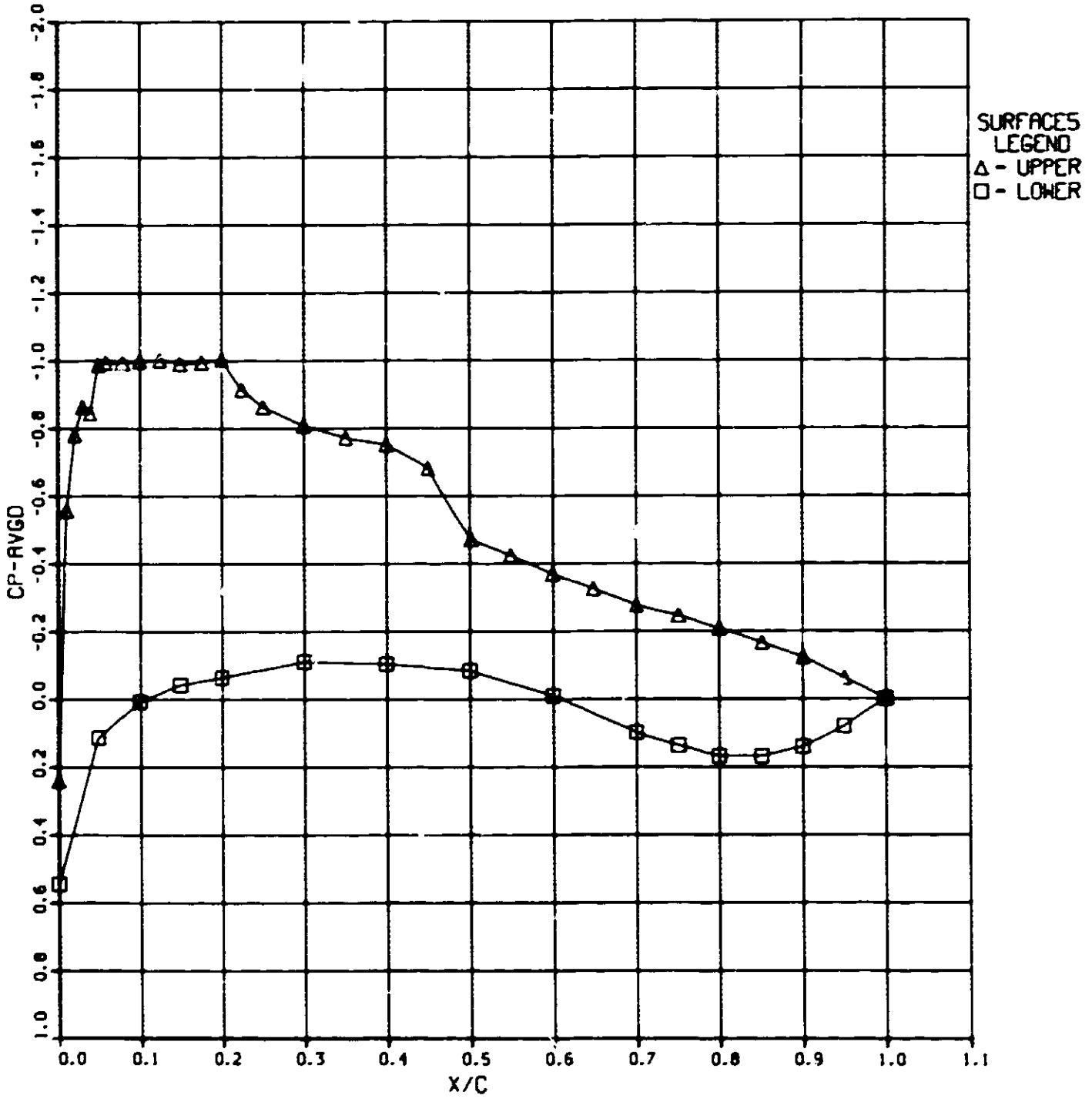

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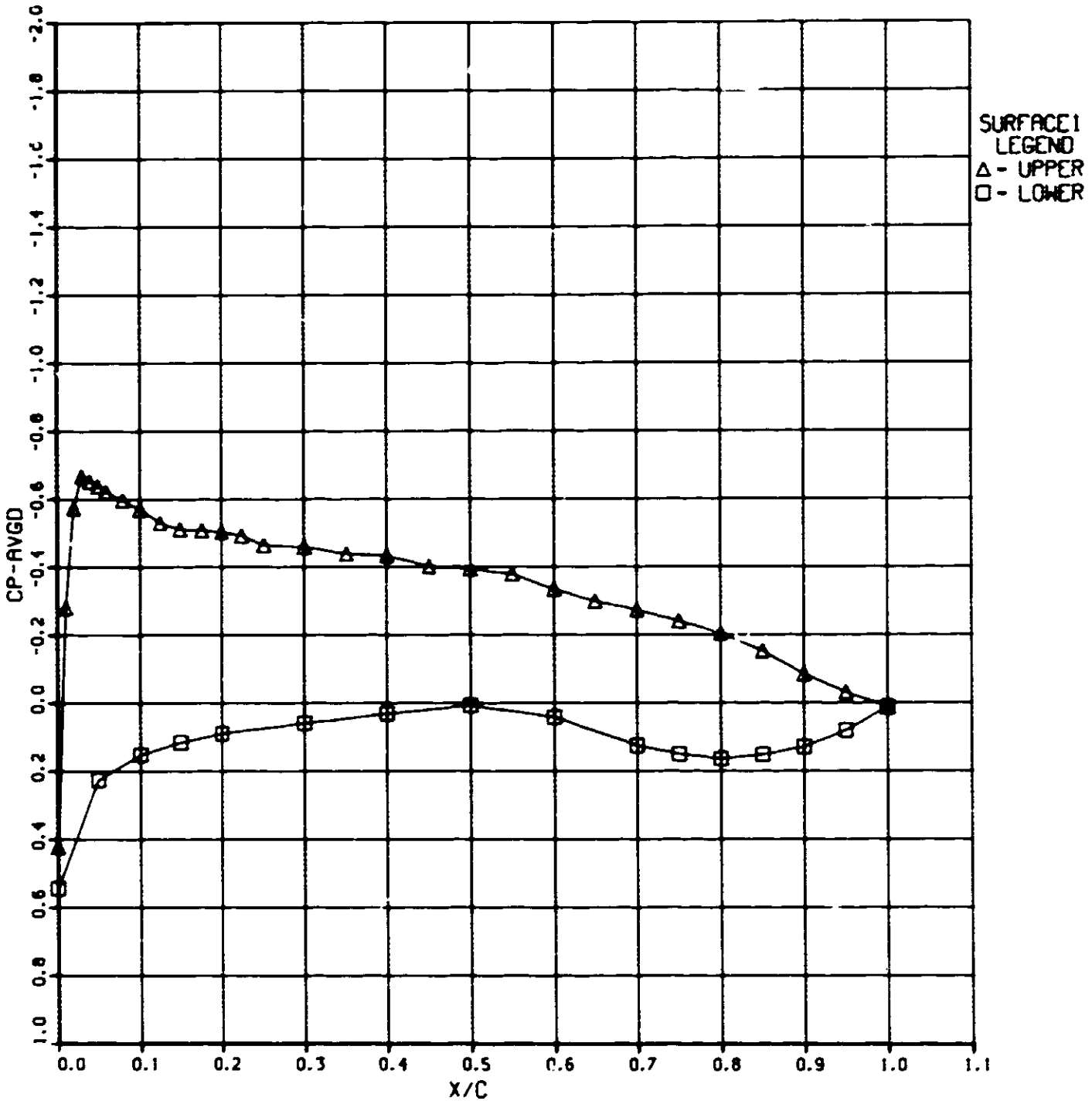

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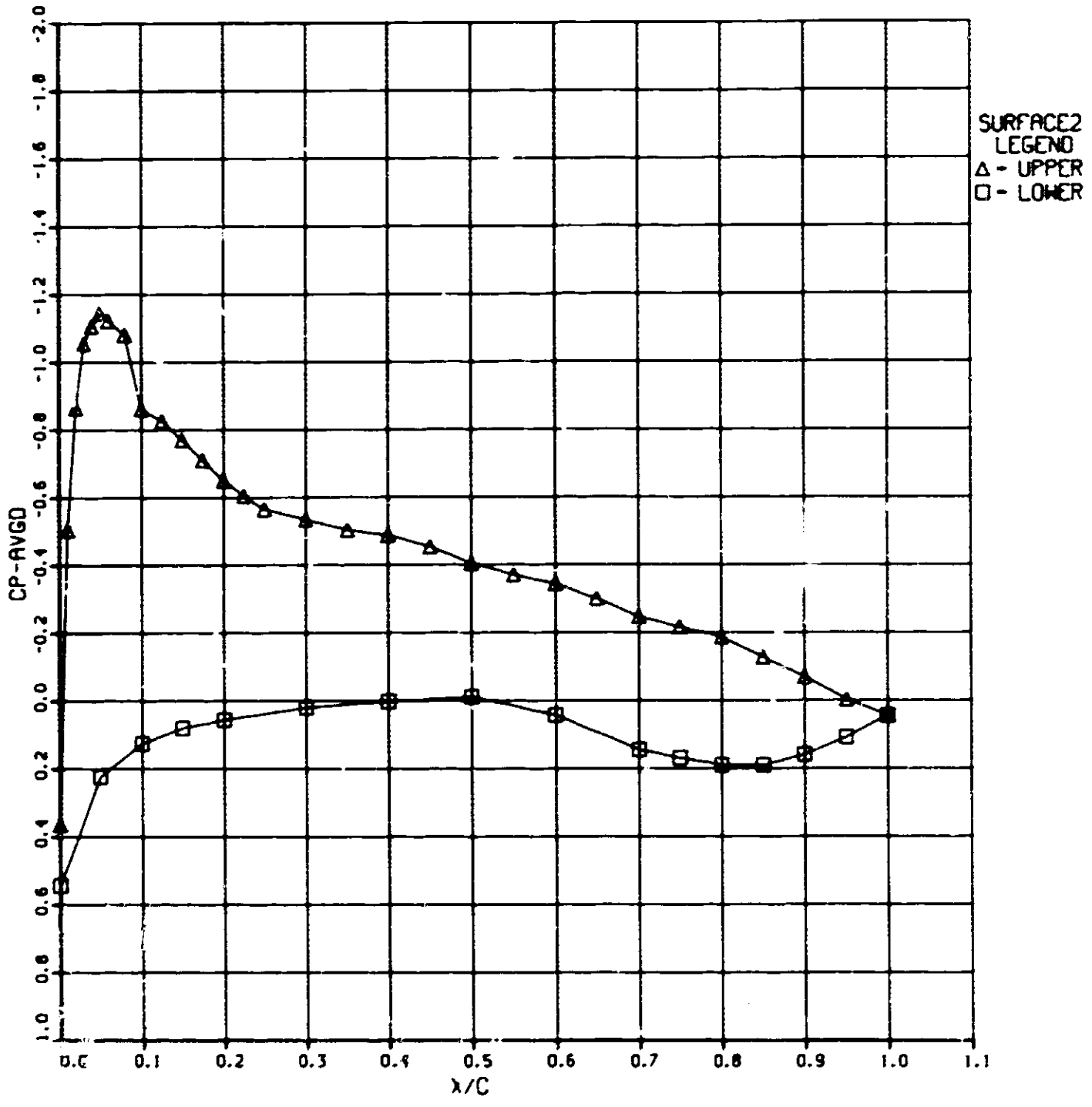

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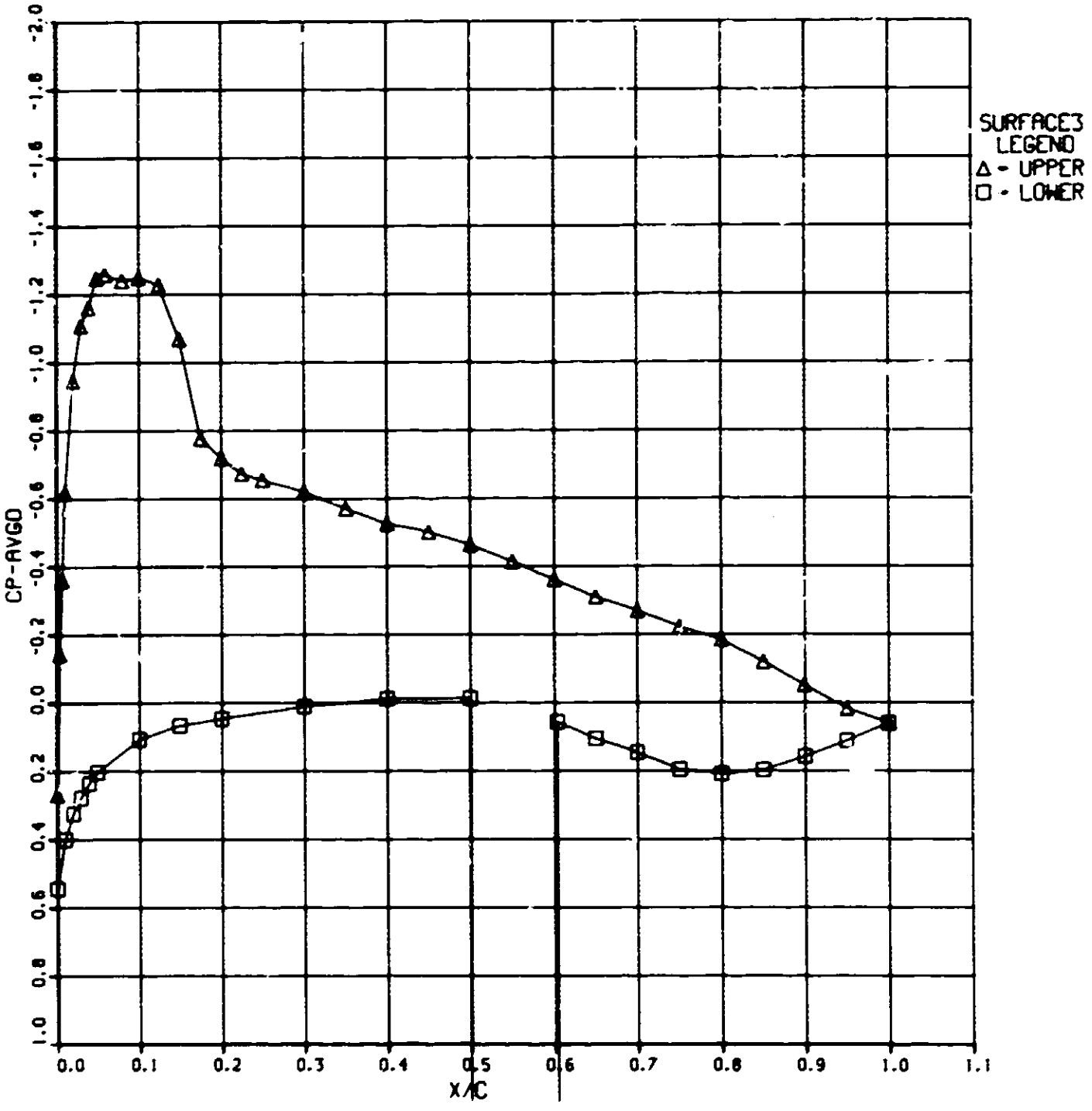

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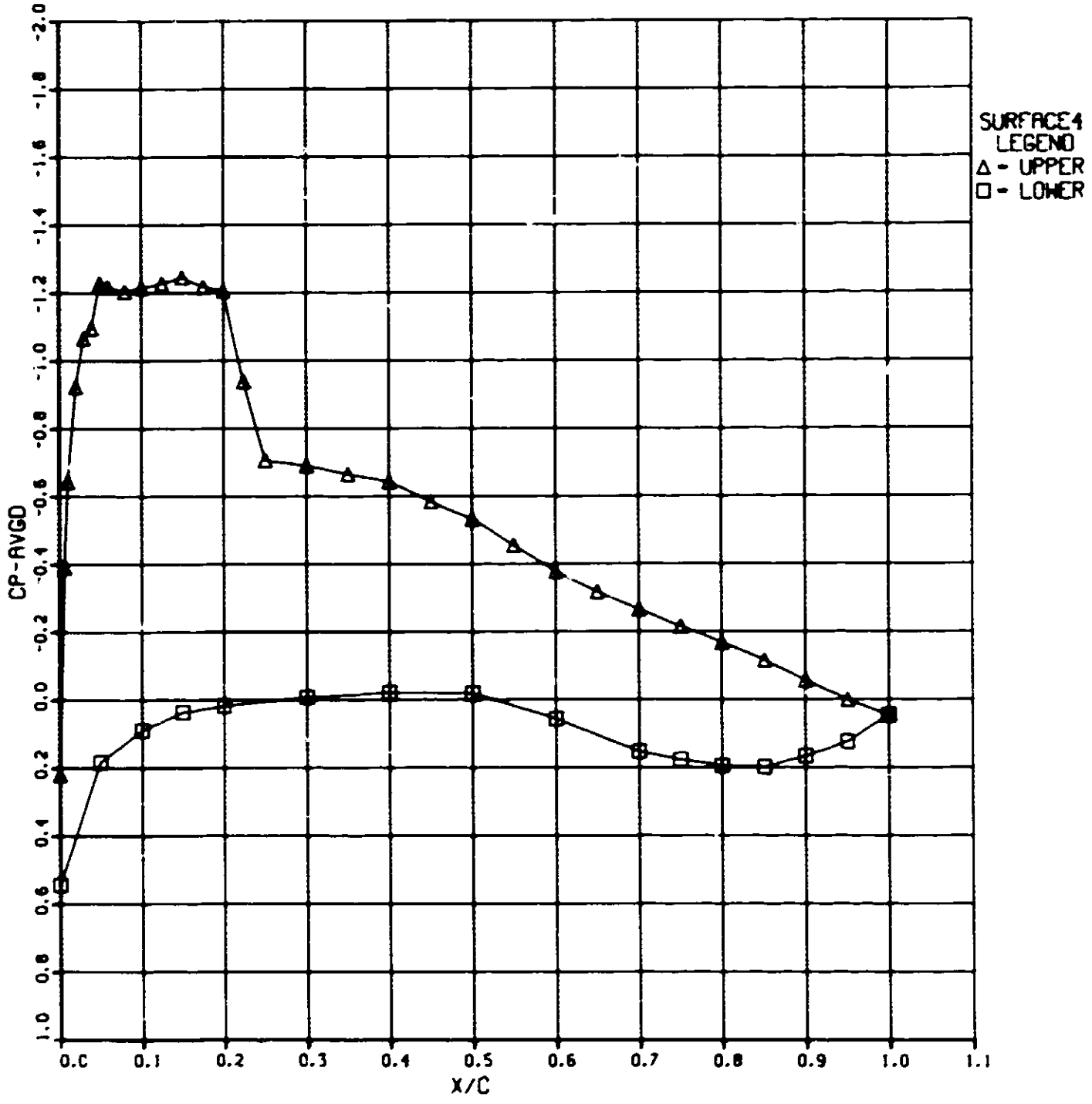

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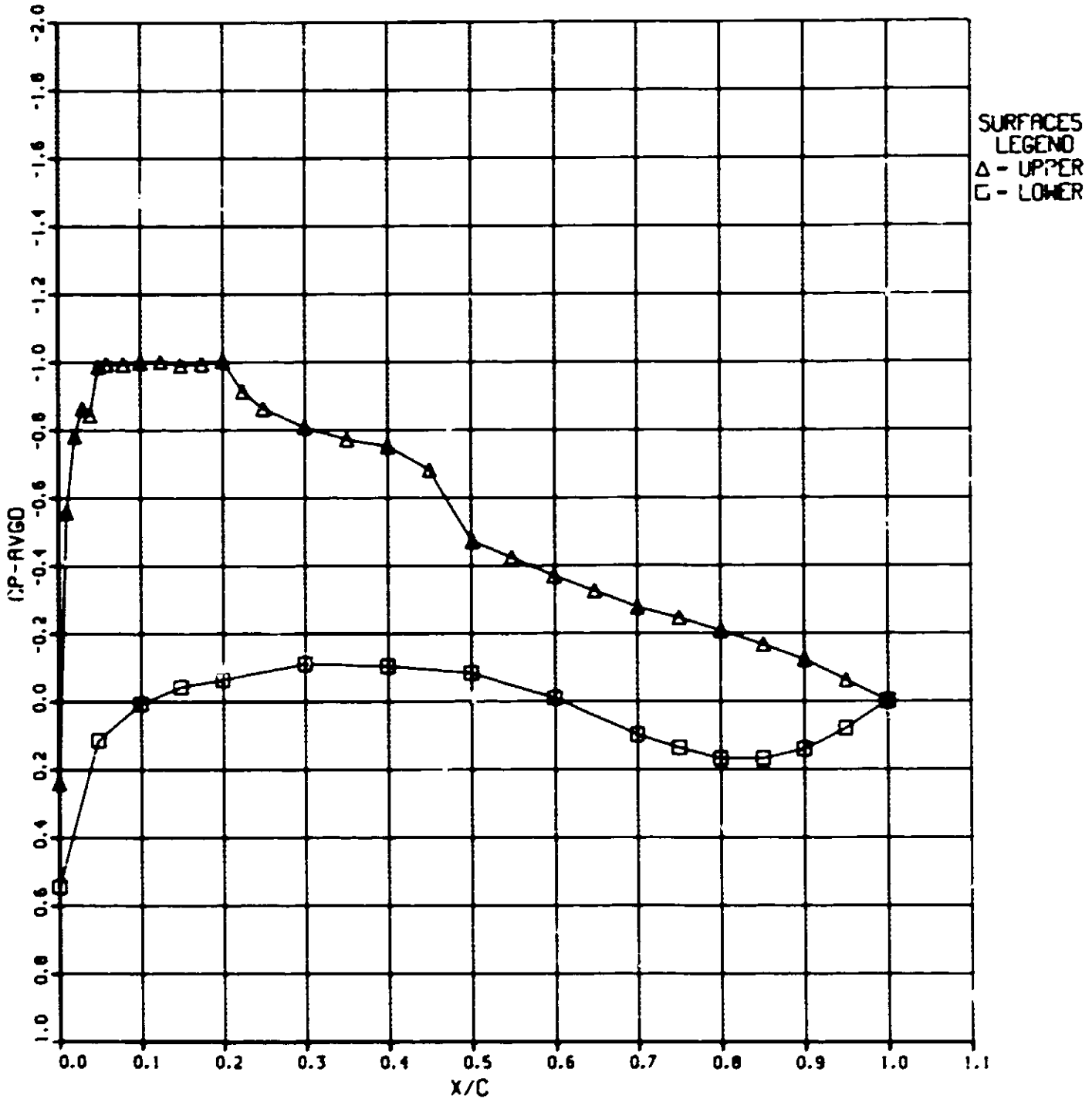

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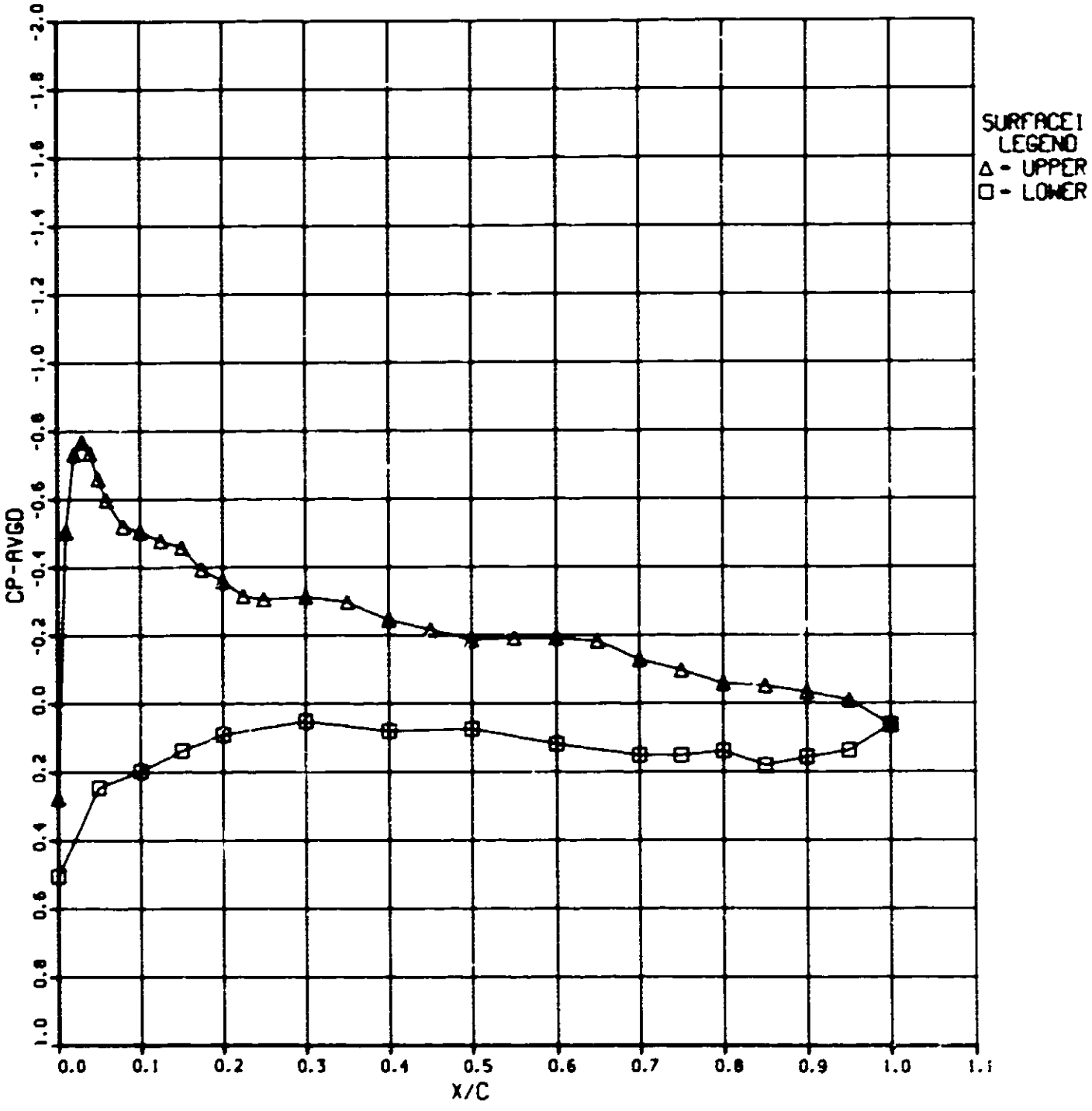

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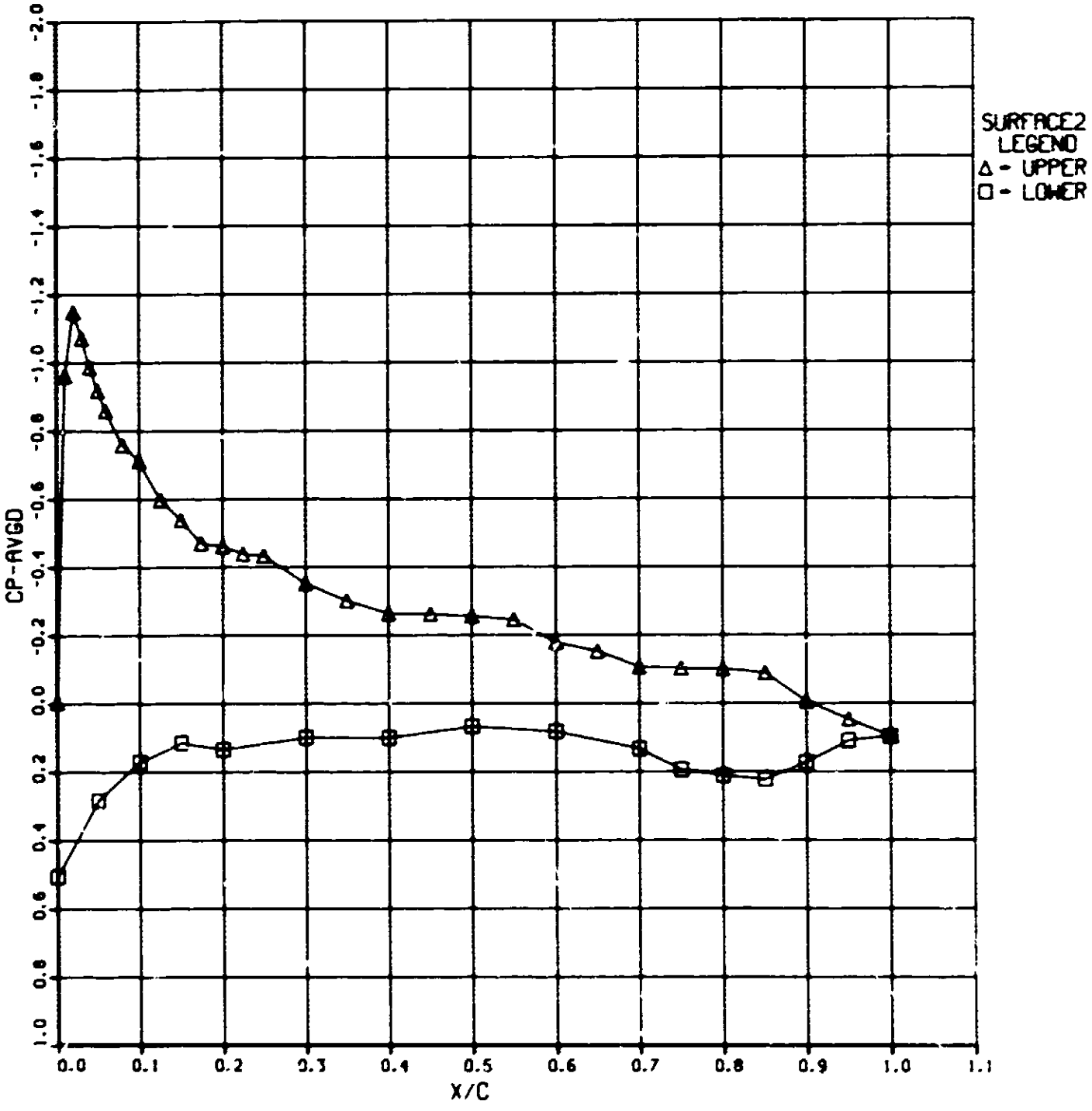

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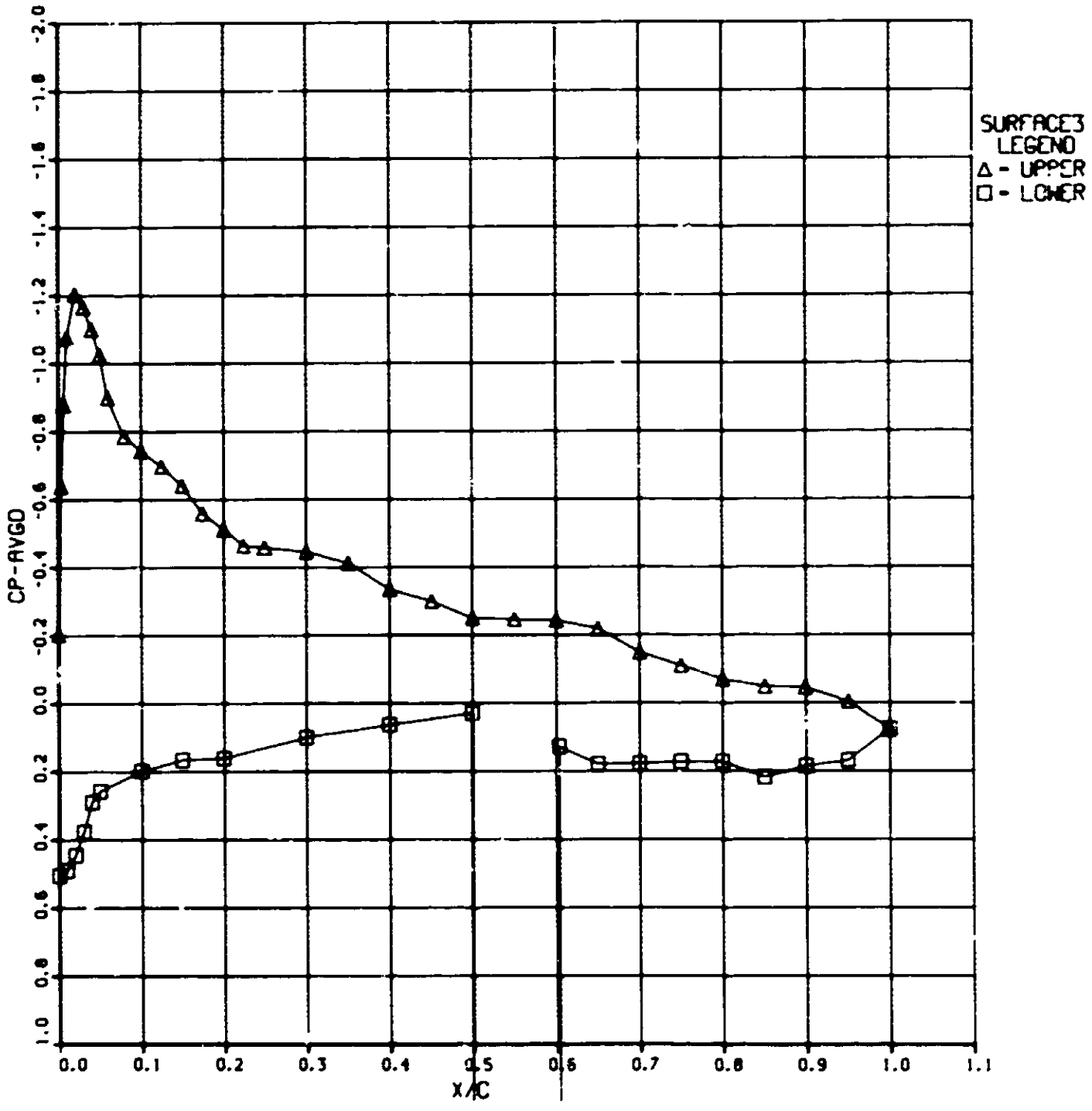

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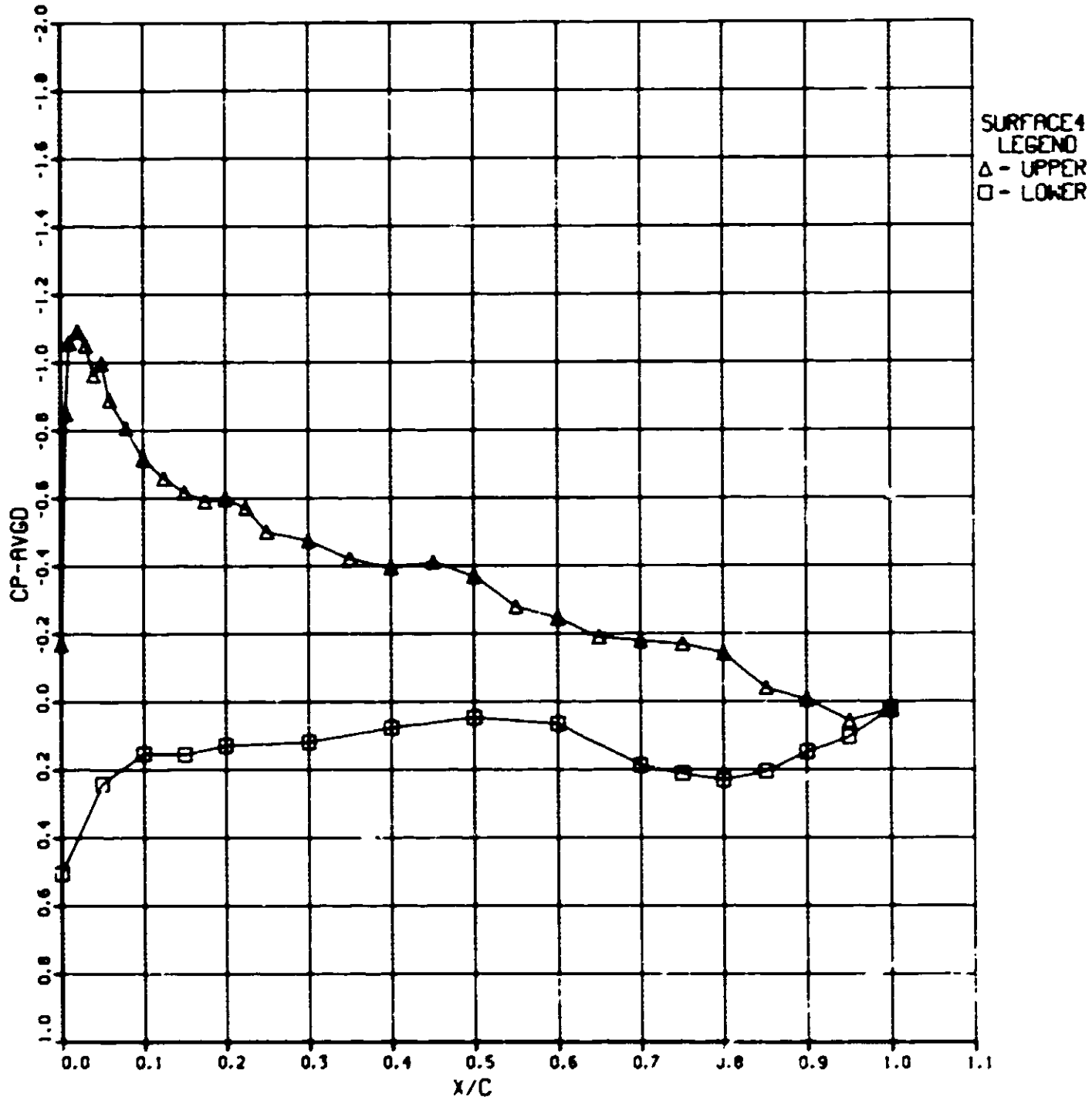

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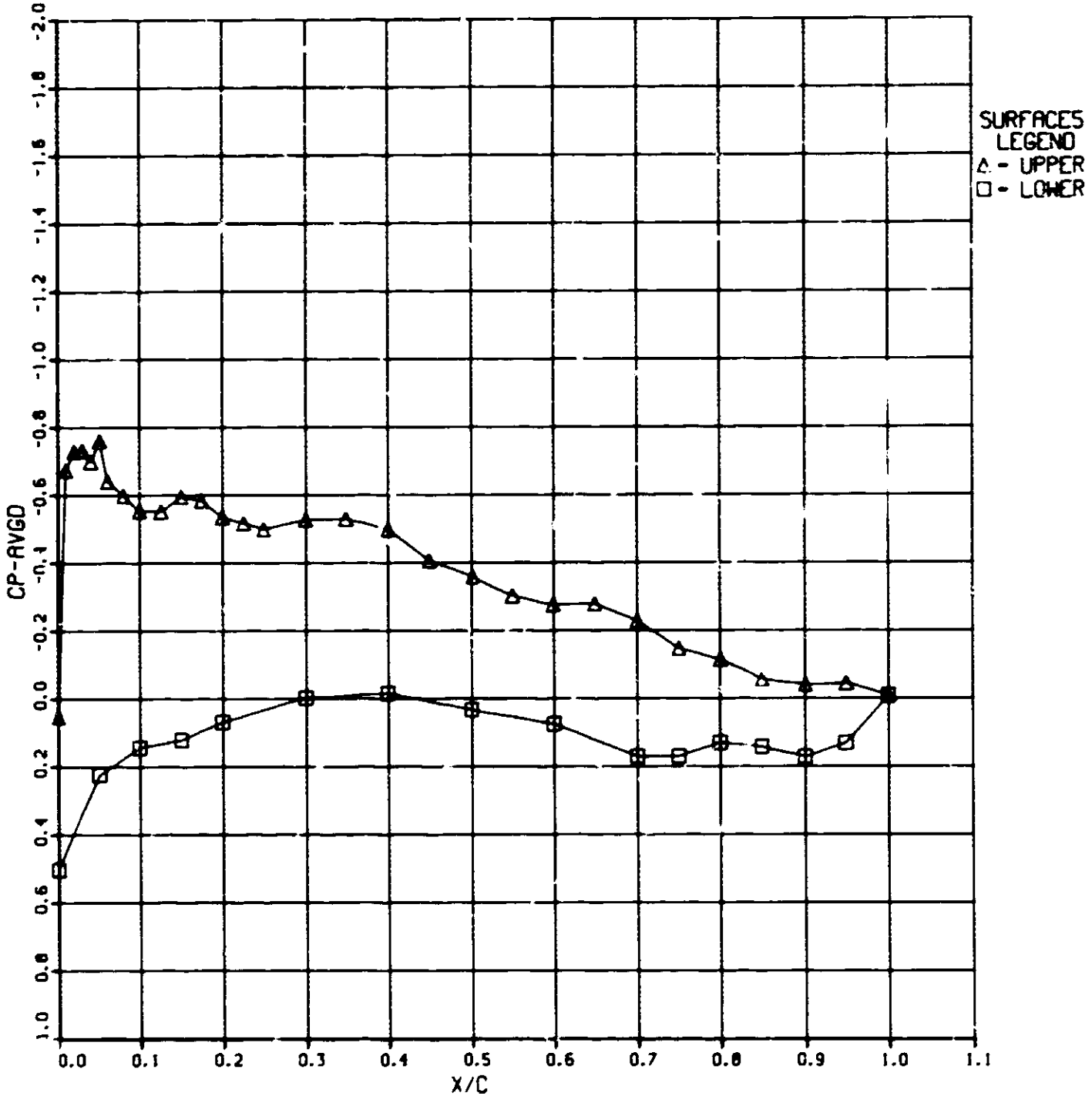

1. Report No.	2. Government Accession No.	3. Recipient's Catalog No.
NASA TM 86683		

4. Title and Subtitle	5. Report Date
PRESSURE-DISTRIBUTION MEASUREMENTS AND OIL-FLOW PATTERNS ON A GENERIC LOW-ASPECT-RATIO WING DESIGNED BY MODERN TRANSONIC, SUPERCRITICAL TECHNOLOGY	September 1985
	6. Performing Organization Code
7. Author(s)	8. Performing Organization Report No.
Earl R. Keener	85131
	10. Work Unit No.
9. Performing Organization Name and Address	
Ames Research Center Moffett Field, CA 94035	11. Contract or Grant No.
	13. Type of Report and Period Covered
12. Sponsoring Agency Name and Address	Technical Report
National Aeronautics and Space Administration Washington, DC 20546	14. Sponsoring Agency Code 505-31-21

15. Supplementary Notes

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16. Abstract

Experimental surface-pressure distributions and oil-flow photographs are presented for a 0.90 m semispan model of NASA/Lockheed Wing C, a generic transonic, supercritical, low-aspect-ratio, highly 3-dimensional configuration. This wing was tested at the design angle of attack of 5° over a Mach number range from 0.25 to 0.96, and a Reynolds number range from 3.4×10^6 to 10×10^6. Pressures were measured with both the tunnel floor and ceiling suction slots open for most of the tests but taped closed for some tests to simulate solid walls. A comparison is made with the measured pressures from a small-scale model in a high Reynolds number facility and with predicted pressures using two three-dimensional (3-D), transonic full-potential-flow wing codes: design code FLO22 (nonconservative) and TWING code (conservative).

At the given design condition, a small region of flow separation occurred. At a Mach number of 0.82 the flow was unseparated and the surface flow-angles were less than 10°, indicating that the boundary-layer flow was not 3-D. Evidence from this study, and from other cited wing studies, indicate that wings that are optimized for mild shock waves and mild pressure-recovery gradients generally have small 3-D boundary-layer flow at design conditions for unseparated flow.

17. Key Words (Suggested by Author(s))	18. Distribution Statement
Wing Transonic Pressure distribution Supercritical Flow visualization	Unlimited Subject category: 02

19. Security Classif. (of this report)	20. Security Classif. (of this page)	21. No. of Pages	22. Price*
Unclassified	Unclassified	75	A04

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