

Issued in Kansas City, Missouri, on August 9, 2006.

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2004-NE-05-AD; Amendment 39-14706; AD 2006-16-06]

RIN 2120-AA64

#### **Airworthiness Directives; General Electric Company (GE) CF6-80 Series Turbofan Engines**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule; request for comments.

**SUMMARY:** The FAA is superseding an existing airworthiness directive (AD) for GE CF6-80 series turbofan engines with certain stage 1 high-pressure turbine (HPT) rotor disks. That AD currently requires an initial inspection as a qualification for the mandatory rework procedures for certain disks, and repetitive inspections only for certain disks for which the rework procedures were not required. That action also requires reworking certain disks before further flight, and removes certain CF6-80E1 series disks from service. This AD requires the same actions but shortens the compliance schedule for HPT disks that have not been previously inspected using AD 2004-04-07, which this AD supersedes. This AD results from a recent report of an uncontained failure of a stage 1 HPT disk. We are issuing this AD to detect and prevent cracks in the bottoms of the dovetail slots that could propagate to failure of the disk and cause an uncontained engine failure.

**DATES:** Effective September 5, 2006. The Director of the Federal Register approved the incorporation by reference of certain publications listed in the regulations as of September 5, 2006. The Director of the Federal Register previously approved the incorporation by reference of certain other publications listed in the regulations as of March 12, 2004 (69 FR 8801, February 26, 2004).

We must receive any comments on this AD by October 17, 2006.

**ADDRESSES:** Use one of the following addresses to submit comments on this ad:

- By mail: Federal Aviation Administration (FAA), New England Region, Office of the Regional Counsel, Attention: Rules Docket No. 2004-NE-05-AD, 12 New England Executive Park, Burlington, MA 01803.

- By fax: (781) 238-7055.

- By e-mail: [9-ane-adcomment@faa.gov](mailto:9-ane-adcomment@faa.gov).

Contact General Electric Company via Lockheed Martin Technology Services, 10525 Chester Road, Suite C, Cincinnati, Ohio 45215, telephone (513) 672-8400, fax (513) 672-8422, for the service information identified in this AD.

You may examine the AD docket, by appointment, at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA. You may examine the service information, by appointment, at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA.

#### **FOR FURTHER INFORMATION CONTACT:**

James Lawrence, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; telephone: (781) 238-7176, fax: (781) 238-7199.

#### **SUPPLEMENTARY INFORMATION:**

On February 13, 2004, we issued AD 2004-04-07, Amendment 39-13488 (69 FR 38; February 26, 2004). That AD requires an initial inspection as a qualification for the mandatory rework procedures for certain disks, and repetitive inspections only for certain disks for which the rework procedures were not required. That action also requires reworking certain disks before further flight. That AD was the result of the manufacturer's investigation and development of a rework procedure to chamfer the aft breakedge of the dovetail slot bottom to reduce stresses. That condition, if not corrected, could result in cracks in the bottoms of the dovetail slots that could propagate to failure of the disk and cause an uncontained engine failure.

#### **Actions Since AD 2004-04-07 Was Issued**

Since AD 2004-04-07 was issued, a CF6-80A turbofan engine, installed on a Boeing 767 airplane, experienced an uncontained stage 1 HPT disk failure on June 2, 2006. The disk failure resulted in a fire and significant damage to the airplane. The event occurred during an on-ground maintenance operation.

#### **Relevant Service Information**

We reviewed and approved the technical contents of the following GE Service Bulletins (SBs) and Alert Service Bulletin (ASB) that describe procedures for removing, inspecting, and reworking certain stage 1 HPT rotor disks:

- SB No. CF6-80E1 S/B 72-0251, dated January 22, 2004;
- SB No. CF6-80A S/B 72-0779, Revision 1, dated January 22, 2004;
- SB No. CF6-80A S/B 72-0788, Revision 3, dated July 20, 2006;
- SB No. CF6-80A S/B 72-0822, dated July 20, 2006;
- ASB No. CF6-80C2 S/B 72-A1026, Revision 2, dated January 22, 2004;
- SB No. CF6-80C2 S/B 72-1089, Revision 3, dated July 20, 2006;
- SB No. CF6-80C2 S/B 72-1217, dated July 20, 2006.

#### **FAA's Determination and Requirements of This AD**

The unsafe condition described previously is likely to exist or develop on other GE CF6-80 series turbofan engines of the same type design. This AD requires rework of the dovetail slot bottom of certain stage 1 rotor disks. The disks must pass an inspection to qualify for the rework. This AD also requires removal from service of certain disks for which the rework procedures were not previously required. This AD also tightens the compliance schedule for HPT disks that have not been previously inspected using AD 2004-04-07. Operators must use the compliance schedule carried forward from AD 2004-04-07 or the new compliance schedule below, whichever occurs first:

- For stage 1 HPT rotor disks with 9,000 or more cycles-since-new (CSN) on the effective date of this AD, within 250 cycles-in-service (CIS) after the effective date of this AD, or by March 31, 2007, whichever occurs first.
- For stage 1 HPT rotor disks with 6,900 or more but fewer than 9,000 CSN on the effective date of this AD, within 500 CIS after the effective date of this AD, or before accumulating 9,250 CSN, or by December 31, 2007, whichever occurs first.
- For stage 1 HPT rotor disk with fewer than 6,900 CSN on the effective date of this AD, before accumulating 7,400 CSN, or by December 31, 2008, whichever occurs first.

This AD also removes from service certain CF6-80E1 series disks. You must use the service information described previously to perform the actions required by this AD.

**FAA's Determination of the Effective Date**

Since an unsafe condition exists that requires the immediate adoption of this AD, we have found that notice and opportunity for public comment before issuing this AD are impracticable, and that good cause exists for making this amendment effective in less than 30 days.

**Comments Invited**

This AD is a final rule that involves requirements affecting flight safety and was not preceded by notice and an opportunity for public comment; however, we invite you to submit any written relevant data, views, or arguments regarding this AD. Send your comments to an address listed under **ADDRESSES**. Include "AD Docket No. 2004-NE-05-D" in the subject line of your comments. If you want us to acknowledge receipt of your mailed comments, send us a self-addressed, stamped postcard with the docket number written on it; we will date-stamp your postcard and mail it back to you. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the rule that might suggest a need to modify it. If a person contacts us verbally, and that contact relates to a substantive part of this AD, we will summarize the contact and place the summary in the docket. We will consider all comments received by the closing date and may amend the AD in light of those comments.

**Examining the AD Docket**

You may examine the AD Docket (including any comments and service information), by appointment, between 8 a.m. and 4:30 p.m., Monday through Friday, except Federal holidays. See **ADDRESSES** for the location.

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority. We are issuing this rulemaking under the authority described in subtitle VII, part A, subpart III, section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

**Regulatory Findings**

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the National Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that the regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a summary of the costs to comply with this AD and placed it in

the AD Docket. You may get a copy of this summary by sending a request to us at the address listed under **ADDRESSES**. Include "AD Docket No. 2004-NE-05-AD" in your request.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**Adoption of the Amendment**

Accordingly, under the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

2. The FAA amends § 39.13 by removing Amendment 39-13488 (69 FR 8801; February 26, 2004), and by adding a new airworthiness directive, Amendment 39-14706, to read as follows:

**2006-16-06 General Electric Company:** Amendment 39-14706. Docket No. 2004-NE-05-AD.

**Effective Date**

(a) This airworthiness directive (AD) becomes effective September 5, 2006.

**Affected ADs**

(b) This AD supersedes AD 2004-04-07 (69 FR 8801; February 26, 2004).

**Applicability**

(c) This AD applies to the General Electric Company (GE) CF6-80 turbofan engine models listed in the following Table 1:

TABLE 1.—APPLICABILITY MODELS, PART NUMBERS, AIRPLANES

Models	Stage 1 high pressure turbine (HPT) rotor disk part numbers (P/Ns)	Engines installed on but not limited to
CF6-80A, CF6-80A1, CF6-80A2, CF6-80A3 ..	9234M67G22/G24/G25/G26, 9362M58G02/G06/G07/G09, 9367M45G02/G04/G09.	Airbus A310 and Boeing 767 airplanes.
CF6-80C2A1, CF6-80C2A2, CF6-80C2A3, CF6-80C2A5, CF6-80C2A8, CF6-80C2A5F, CF6-80C2B1, CF6-80C2B2, CF6-80C2B4, CF6-80C2B6, CF6-80C2B1F, CF6-80C2B2F, CF6-80C2B4F, CF6-80C2B5F, CF6-80C2B6F, CF6-80C2B6FA, CF6-80C2B7F, CF6-80C2D1F.	1862M23G01, 9392M23G10/G12/G21, 1531M84G02/G06/G08/G10/G12.	Airbus A300, A310, Boeing 747, 767, and McDonnell Douglas MD11 airplanes.
CF6-80E1A2, CF6-80E1A4 .....	1639M41P04 .....	Airbus A330 airplanes.

These engines are installed on, but not limited to, the airplanes listed in Table 1 of this AD.

**Unsafe Condition**

(d) This AD results from a recent report of an uncontained failure of a stage 1 HPT disk.

The actions specified in this AD are intended to detect and prevent cracks in the bottoms of the dovetail slots that could propagate to

failure of the disk and cause an uncontained engine failure.

**Compliance**

(e) You are responsible for having the actions required by this AD performed within the compliance times specified unless the actions have already been done.

*CF6-80A, -80A1, -80A2, and -80A3 Engines*

Stage 1 HPT Rotor Disks, P/N 9362M58G09, With Chamfered Breakedges

(f) At the next piece-part exposure, for stage 1 HPT rotor disks, P/N 9362M58G09, with serial numbers (SNs) listed in Table 2 of this AD, do the following, unless already done using superseded AD 2004-04-07:

**TABLE 2.—SNS OF CF6-80A SERIES STAGE 1 HPT ROTOR DISK P/N 9362M58G09—WITH CHAMFERED BREAKEDGES**

- GWN03RD7
- GWN03TKG
- GWN03TKH
- GWN03TKJ
- GWN03W3M
- GWN03W3N
- GWN03W3R
- GWN042J3
- GWN04FW2
- GWN04FW3
- GWN04FW4

**TABLE 2.—SNS OF CF6-80A SERIES STAGE 1 HPT ROTOR DISK P/N 9362M58G09—WITH CHAMFERED BREAKEDGES—Continued**

- GWN04FW5
- GWN04HOM
- GWN04HRA
- GWN04HRD
- GWN04HRE
- GWN04HRF
- GWN04HRG
- GWN04HRH
- GWN04K8N
- GWN04M9J
- GWN04M9K
- GWN04M9L
- GWN04M9M
- GWN04M9R
- GWN04M9T
- GWN04M9W

(1) Visually inspect the rotor disks for the presence of a chamfer on the aft breakedges of the dovetail slot bottoms. Use paragraph 3.A. of GE Service Bulletin (SB) No. CF6-80A S/B 72-0822, dated July 20, 2006, to do the inspection.

(2) For disks that have the chamfered breakedges, re-mark, fluorescent penetrant inspect (FPI), and eddy current inspect (ECI) the rotor disk. Use paragraph 3.A.(1) of the Accomplishment Instructions of GE SB No. CF6-80A S/B 72-0822, dated July 20, 2006,

to re-mark and inspect the rotor disk and remove from service as necessary.

(3) For disks that do not have the chamfered breakedges, remove the disk from service. Use paragraph 3.A(2) of the Accomplishment Instructions of GE SB No. CF6-80A S/B 72-0822, dated July 20, 2006.

Stage 1 HPT Rotor Disks, P/Ns 9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, and 9362M58G09 with SNs not listed in Table 2 of this AD

(g) For stage 1 HPT rotor disks, P/Ns 9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, and 9362M58G09 with SNs not listed in Table 2 of this AD, inspect, rework, and re-mark the disks using paragraphs 3.A.(1) through 3.A.(2) of Accomplishment Instructions of GE SB No. CF6-80A S/B 72-0788, Revision 3, dated July 20, 2006, at the following, unless already done using superseded AD 2004-04-07:

(1) For both new and used stage 1 HPT rotor disks not installed in engines, inspect, rework, re-mark, and remove from service as necessary before further flight.

(2) For stage 1 HPT rotor disks that have been inspected using any version of GE SB No. CF6-80A S/B 72-0779, inspect, rework, re-mark, and remove from service as necessary at the next Engine Shop Visit (ESV) using the compliance times in the following Table 3:

**TABLE 3.—COMPLIANCE TIMES FOR INSPECTION AND REWORK OF CF6-80A SERIES STAGE 1 HPT ROTOR DISKS, P/Ns 9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, AND 9362M58G09 WITH SNs NOT LISTED IN TABLE 2 OF THIS AD—PREVIOUSLY INSPECTED**

Stage 1 HPT rotor disk cycles-since-last-inspection (CSLI) on March 12, 2004 (effective date of superseded AD 2004-04-07)	Compliance time for inspection and rework
(i) More than 1,500 CSLI .....	At the next ESV after March 12, 2004 (effective date of superseded AD 2004-04-07), but not to exceed 4,500 CSLI.
(ii) 1,500 CSLI or fewer .....	At the next ESV after March 12, 2004 (effective date of superseded AD 2004-04-07), but not to exceed 3,500 CSLI.

(3) For stage 1 HPT rotor disks which have not been inspected using any version of GE SB No. CF6-80A S/B 72-0779, inspect,

rework, re-mark, and remove from service as necessary using the following Table 4 or

Table 4A compliance times, whichever occurs first:

**TABLE 4.—COMPLIANCE TIMES FOR INSPECTION AND REWORK OF CF6-80A SERIES STAGE 1 HPT ROTOR DISKS, P/Ns 9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, AND 9362M58G09 WITH SNs NOT LISTED IN TABLE 2 OF THIS AD—NOT PREVIOUSLY INSPECTED**

Stage 1 HPT rotor disk cycles-since-new (CSN) on the effective date of this AD	Compliance time for inspection and rework
(i) 9,000 or more CSN .....	Within 250 cycles-in-service (CIS) after the effective date of this AD, or by March 31, 2007, whichever occurs first.
(ii) 6,900 or more but fewer than 9,000 CSN .....	Within 500 CIS after the effective date of this AD, but before accumulating 9,250 CSN, or by December 31, 2007, whichever occurs first.
(iii) Fewer than 6,900 CSN .....	Before accumulating 7,400 CSN, or by December 31, 2008, whichever occurs first.

TABLE 4A.—COMPLIANCE TIMES FOR INSPECTION AND REWORK OF CF6–80A SERIES STAGE 1 HPT ROTOR DISKS, P/Ns 9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, AND 9362M58G09 WITH SNs NOT LISTED IN TABLE 2 OF THIS AD—NOT PREVIOUSLY INSPECTED

Stage 1 HPT rotor disk CSN on March 12, 2004 (effective date of superseded AD 2004–04–07)	Compliance time for inspection and rework
(i) 10,000 or more CSN .....	At the next ESV or within 1,000 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first.
(ii) 5,000 or more CSN but fewer than 10,000 CSN .....	At the next ESV or within 2,400 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first, but before accumulating 11,000 CSN.
(iii) Fewer than 5,000 CSN .....	At the next ESV or within 3,500 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first, but before accumulating 7,400 CSN.

Stage 1 HPT Rotor Disks, P/N 9367M45G02

(h) For stage 1 HPT rotor disks, P/N 9367M45G02, remove the disk from service at the following times:

(1) For stage 1 HPT rotor disks not installed in engines, remove from service before further flight.

(2) For stage 1 HPT rotor disks that have been inspected before the effective date of this AD using any version of GE SB No. CF6–80A S/B 72–0779, and had more than zero CSN at the time of that inspection, remove from service at next ESV.

(3) For stage 1 HPT rotor disks that have not been inspected, or were only inspected with zero CSN before the effective date of this AD using any version of GE SB No. CF6–80A S/B 72–0779, remove from service using the following Table 5 or Table 5A compliance times, whichever occurs first:

TABLE 5.—COMPLIANCE TIMES FOR REMOVAL OF CF6–80A SERIES STAGE 1 HPT ROTOR DISKS, P/N 9367M45G02—NOT PREVIOUSLY INSPECTED

Stage 1 HPT rotor disk CSN on the effective date of this AD	Compliance time for removal
(i) 9,000 or more CSN .....	Within 250 CIS after the effective date of this AD, or by March 31, 2007, whichever occurs first.
(ii) 6,900 or more but fewer than 9,000 CSN .....	Within 500 CIS after the effective date of this AD, but before accumulating 9,250 CSN, or by December 31, 2007, whichever occurs first.
(iii) Fewer than 6,900 CSN .....	Before accumulating 7,400 CSN, or by December 31, 2008, whichever occurs first.

TABLE 5A.—COMPLIANCE TIMES FOR REMOVAL OF CF6–80A SERIES STAGE 1 HPT ROTOR DISKS, P/N 9367M45G02—NOT PREVIOUSLY INSPECTED

Stage 1 HPT rotor disk CSN on March 12, 2004 (effective date of superseded AD 2004–04–07)	Compliance time for removal
(i) 10,000 or more CSN .....	At the next ESV or within 1,000 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first.
(ii) 5,000 or more CSN but fewer than 10,000 CSN .....	At the next ESV or within 2,400 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first, but before accumulating 11,000 CSN.
(iii) Fewer than 5,000 CSN .....	At the next ESV or within 3,500 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first, but before accumulating 7,400 CSN.

CF6–80C2 Series Engines

Stage 1 HPT Rotor Disks, P/N 1531M84G10, With Chamfered Breakedges, Group 1

(i) At the next piece-part exposure, for stage 1 HPT rotor disks, P/N 1531M84G10, with SNs listed in Table 6 (Group 1) of this AD, do the following, unless already done using superseded AD 2004–04–07:

TABLE 6.—SNs OF CF6–80C2 SERIES STAGE 1 HPT ROTOR DISKS, P/N 1531M84G10, WITH CHAMFERED BREAKEDGES, GROUP 1

- GWN03111
- GWN03114
- GWN031N2

TABLE 6.—SNs OF CF6–80C2 SERIES STAGE 1 HPT ROTOR DISKS, P/N 1531M84G10, WITH CHAMFERED BREAKEDGES, GROUP 1—Continued

- GWN031N3
- GWN031N4
- GWN031N5
- GWN031N6
- GWN031N7
- GWN031N8
- GWN031N9
- GWN031NA
- GWN031NC
- GWN032G1
- GWN032G2
- GWN032G3
- GWN032G4
- GWN032G5

TABLE 6.—SNs OF CF6–80C2 SERIES STAGE 1 HPT ROTOR DISKS, P/N 1531M84G10, WITH CHAMFERED BREAKEDGES, GROUP 1—Continued

- GWN032G6
- GWN032G7
- GWN032G8
- GWN032G9
- GWN032GE
- GWN0335P
- GWN0335R
- GWN033C5
- GWN034KR
- GWN034KT
- GWN03501
- GWN0350M
- GWN0350N
- GWN0350P

TABLE 6.—SNS OF CF6–80C2 SERIES  
STAGE 1 HPT ROTOR DISKS, P/N  
1531M84G10, WITH CHAMFERED  
BREAKEDGES, GROUP 1—Continued

GWN0350R  
GWN0350T  
GWN0350W  
GWN035M5  
GWN035M6  
GWN035M7  
GWN035M8  
GWN035M9  
GWN035MA  
GWN035MC  
GWN035MD  
GWN035TH  
GWN035TJ  
GWN035TK  
GWN035TL  
GWN035TM  
GWN03699  
GWN0369A  
GWN0369C  
GWN0369D  
GWN0369E  
GWN0369G  
GWN0369H  
GWN0369J  
GWN036JG  
GWN036JH  
GWN036JJ  
GWN036JK  
GWN036JL  
GWN036JM  
GWN036JN  
GWN03752  
GWN03753  
GWN03754  
GWN03755  
GWN03756  
GWN03757  
GWN03759  
GWN0375A  
GWN0375C  
GWN0375D  
GWN0375E  
GWN037H2  
GWN03981  
GWN03982  
GWN03983  
GWN03984  
GWN03985  
GWN03986  
GWN03987  
GWN03988  
GWN03989  
GWN0398A  
GWN0398C  
GWN039PF  
GWN039PG  
GWN039PH  
GWN039PJ  
GWN039PK  
GWN039PL  
GWN039PM  
GWN039PN  
GWN03A4J  
GWN03A4K  
GWN03A4L  
GWN03A4M  
GWN03A4N  
GWN03A4P  
GWN03A4R  
GWN03A4T  
GWN03A4W

TABLE 6.—SNS OF CF6–80C2 SERIES  
STAGE 1 HPT ROTOR DISKS, P/N  
1531M84G10, WITH CHAMFERED  
BREAKEDGES, GROUP 1—Continued

GWN03C12  
GWN03C13  
GWN03C14  
GWN03CA0  
GWN03DC9  
GWN03DCA  
GWN03DCC  
GWN03DCD  
GWN03DCE  
GWN03DCF  
GWN03DCG  
GWN03DCH  
GWN03DCJ  
GWN03DCK  
GWN03DCL  
GWN03DCM  
GWN03DCN  
GWN03DCP  
GWN03DCR  
GWN03DME  
GWN03DMF  
GWN03ER7  
GWN03ER8  
GWN03ER9  
GWN03ERA  
GWN03FTN  
GWN03FTP  
GWN03FTR  
GWN03FTT  
GWN03FTW  
GWN03FW0  
GWN03H56  
GWN03H57  
GWN03H58  
GWN03HTL  
GWN03HTM  
GWN03HTN  
GWN03HTP  
GWN03HTR  
GWN03HTT  
GWN03J8T  
GWN03J8W  
GWN03J91  
GWN03J92  
GWN03JNN  
GWN03JNP  
GWN03K3C  
GWN03K3D  
GWN03K3F  
GWN03K3G  
GWN03K3H  
GWN03K3K  
GWN03K3L  
GWN03K3M  
GWN03K3N  
GWN03K3T  
GWN03K3W  
GWN03K40  
GWN03K7R  
GWN03KR1  
GWN03KR3  
GWN03KR4  
GWN03KR6  
GWN03KR7  
GWN03KR8  
GWN03KRC  
GWN03L2D  
GWN03L2E  
GWN03L2F  
GWN03LNF  
GWN03LNJ

TABLE 6.—SNS OF CF6–80C2 SERIES  
STAGE 1 HPT ROTOR DISKS, P/N  
1531M84G10, WITH CHAMFERED  
BREAKEDGES, GROUP 1—Continued

GWN03LNK  
GWN03M88  
GWN03M8C  
GWN03M8E  
GWN03M8J  
GWN03M8K  
GWN03NHN  
GWN03NHP  
GWN03NHR  
GWN03R74  
GWN03R76  
GWN03R78  
GWN03R7E  
GWN03R7F  
GWN03R9G  
GWN03R9H  
GWN03R9M  
GWN03R9P  
GWN03R9T  
GWN03RA2  
GWN03RA3  
GWN03RA5  
GWN03RA8  
GWN03RPA  
GWN03RPC  
GWN03RPD  
GWN04026  
GWN0402A  
GWN0402F  
GWN0402L  
GWN040R5  
GWN04189  
GWN0418A  
GWN0418D  
GWN0418E  
GWN0418F  
GWN0418H  
GWN0418J  
GWN0418L  
GWN0418N  
GWN0418R  
GWN04366  
GWN044DP  
GWN0454H  
GWN0454M  
GWN0454N  
GWN045T0  
GWN045T2  
GWN045T8  
GWN045TD  
GWN045TG  
GWN04722  
GWN04729  
GWN047LK  
GWN048CD  
GWN048CF  
GWN048CH  
GWN048CJ  
GWN048CK  
GWN049GJ  
GWN049M8  
GWN049M9  
GWN04AER  
GWN04ALR  
GWN04AM1  
GWN04CGJ  
GWN04CGN  
GWN04CGT  
GWN04CGW  
GWN04CH3  
GWN04CH5

TABLE 6.—SNS OF CF6–80C2 SERIES STAGE 1 HPT ROTOR DISKS, P/N 1531M84G10, WITH CHAMFERED BREAKEDGES, GROUP 1—Continued

GWN04CH8  
GWN04CH9  
GWN04D52  
GWN04D54  
GWN04D56  
GWN04D57  
GWN04D58  
GWN04D59  
GWN04DPW  
GWN04E9K  
GWN04E9L  
GWN04E9M  
GWN04EMA  
GWN04EMK  
GWN04EML  
GWN04EMM  
GWN04FTL  
GWN04FTM  
GWN04FTN

(1) Visually inspect the rotor disks for the presence of a chamfer on the aft breakedges of the dovetail slot bottoms. Use paragraph 3.A. of GE SB No. CF6–80C2 S/B 72–1217, dated July 20, 2006, to do the inspection.

(2) For disks that have the chamfered breakedges, re-mark, FPI, and ECI the rotor disk. Use paragraph 3.A.(1) of the Accomplishment Instructions of GE SB No. CF6–80C2 S/B 72–1217, dated July 20, 2006, to re-mark and inspect the rotor disk, and remove from service as necessary.

(3) For disks that do not have the chamfered breakedges, remove the disk from service. Use paragraph 3.A.(4) of the Accomplishment Instructions of GE SB No. CF6–80C2 S/B 72–1217, dated July 20, 2006.

*CF6–80C2 Series Engines*

Stage 1 HPT Rotor Disks, P/N 1531M84G10, With Chamfered Breakedges, Group 2

(j) For stage 1 HPT rotor disks, P/N 1531M84G10, with SNs listed in Table 6A of this AD, with chamfered breakedges, (Group 2):

(1) With more than 6,900 CSN, perform paragraphs (j)(3) through (j)(5) as applicable, at the next ESV, but within 500 CIS after the effective date of this AD, unless already done using superseded AD 2004–04–07.

(2) With 6,900 CSN or fewer, perform paragraphs (j)(3) through (j)(5) as applicable, at the next ESV, but before accumulating 7,400 CSN, unless already done using superseded AD 2004–04–07.

TABLE 6A.—SNS OF CF6–80C2 SERIES STAGE 1 HPT ROTOR DISKS, P/N 1531M84G10, WITH CHAMFERED BREAKEDGES, GROUP 2

GWN03J90  
GWN03K3R  
GWN03K6J  
GWN03K7T  
GWN03KR2  
GWN03KR5  
GWN03KRA  
GWN03KRD

TABLE 6A.—SNS OF CF6–80C2 SERIES STAGE 1 HPT ROTOR DISKS, P/N 1531M84G10, WITH CHAMFERED BREAKEDGES, GROUP 2—Continued

GWN03M89  
GWN03M8D  
GWN03M8F  
GWN03NHT  
GWN03R73  
GWN03R75  
GWN03R77  
GWN03R79  
GWN03R7A  
GWN03R7C  
GWN03R7D  
GWN03R7G  
GWN03R7H  
GWN03R9J  
GWN03R9K  
GWN03R9L  
GWN03R9N  
GWN03R9R  
GWN03R9W  
GWN03RA0  
GWN03RA1  
GWN03RA4  
GWN03RA6  
GWN03RA7  
GWN03RP7  
GWN03RP9  
GWN03RPE  
GWN03RPF  
GWN03RPG  
GWN04027  
GWN04028  
GWN04029  
GWN0402E  
GWN0402G  
GWN0402H  
GWN0402J  
GWN0402K  
GWN0402M  
GWN0402N  
GWN0402P  
GWN0418C  
GWN0418G  
GWN0418K  
GWN0418M  
GWN0418P  
GWN0418T  
GWN0418W  
GWN04190  
GWN04191  
GWN0454E  
GWN0454F  
GWN0454G  
GWN0454J  
GWN0454K  
GWN0454L  
GWN045T1  
GWN045T3  
GWN045T4  
GWN045T5  
GWN045T6  
GWN045T7  
GWN045T9  
GWN045TA  
GWN045TC  
GWN045TE  
GWN045TF  
GWN045TH  
GWN046F6  
GWN046F7

TABLE 6A.—SNS OF CF6–80C2 SERIES STAGE 1 HPT ROTOR DISKS, P/N 1531M84G10, WITH CHAMFERED BREAKEDGES, GROUP 2—Continued

GWN046F8  
GWN04726  
GWN047LG  
GWN047LH  
GWN047LJ  
GWN047LL  
GWN048CG  
GWN048CM  
GWN048CN  
GWN048CP  
GWN048CR  
GWN049GH  
GWN049GK  
GWN049JL  
GWN049JM  
GWN049M7  
GWN04AEP  
GWN04AET  
GWN04ALT  
GWN04ALW  
GWN04AM0  
GWN04AM2  
GWN04AM3  
GWN04AM4  
GWN04CGL  
GWN04CHA  
GWN04CHC  
GWN04D55  
GWN04DR4  
GWN04DR9  
GWN04DRE  
GWN04DRJ  
GWN04E9N  
GWN04EM5  
GWN04F8N  
GWN04F8P  
GWN04FTJ

(3) Visually inspect the rotor disks for the presence of a chamfer on the aft breakedges of the dovetail slot bottoms. Use paragraph 3.A. of GE SB No. CF6–80C2 S/B 72–1217, dated July 20, 2006, to do the inspection.

(4) For disks that have the chamfered breakedges, re-mark, FPI, and ECI the rotor disk. Use paragraph 3.A.(2) of the Accomplishment Instructions of GE SB No. CF6–80C2 S/B 72–1217, dated July 20, 2006, to re-mark and inspect the rotor disk, and remove from service as necessary.

(5) For disks that do not have the chamfered breakedges, remove the disk from service. Use paragraph 3.A.(4) of the Accomplishment Instructions of GE SB No. CF6–80C2 S/B 72–1217, dated July 20, 2006.

*CF6–80C2 Series Engines*

Stage 1 HPT Rotor Disks, P/N 1531M84G12, With Chamfered Breakedges

(k) For stage 1 HPT rotor disks, P/N 1531M84G12, with SNs listed in Table 6B of this AD, with chamfered breakedges:

(1) With more than 6,900 CSN, perform paragraph (k)(3) at the next ESV, but not to exceed 500 cycles after the effective date of this AD.

(2) With 6,900 CSN or fewer, perform paragraph (k)(3) at the next ESV, but before accumulating 7,400 CSN.

**TABLE 6B.—SNS OF CF6–80C2 SERIES STAGE 1 HPT ROTOR DISKS, P/N 1531M84G12, WITH CHAMFERED BREAKEDGES**

GWN04CH6  
GWN04G5H  
GWN04M03

(3) FPI and ECI the rotor disk. Use paragraph 3.A.(3) of the Accomplishment Instructions of GE SB No. CF6–80C2 S/B 72–1217, dated July 20, 2006, to re-mark and

inspect the rotor disk, and remove from service as necessary.  
Stage 1 HPT Rotor Disks, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, and 1531M84G10 with SNs not listed in Table 6 and Table 6A of this AD

(l) For stage 1 HPT rotor disks, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, and 1531M84G10 with SNs not listed in Table 6 and Table 6A of this AD, inspect, rework, and re-mark the disks using paragraphs 3.A.(1) through 3.A.(2) of Accomplishment Instructions of GE SB No. CF6–80C2 S/B 72–1089, Revision 3, dated

July 20, 2006, at the following, unless already done using superseded AD 2004–04–07:

(1) For both new and used stage 1 HPT rotor disks not installed in engines, inspect, rework, re-mark, and remove from service as necessary before further flight.

(2) For stage 1 HPT rotor disks that have been inspected before March 12, 2004 (effective date of superseded AD 2004–04–07) using GE ASB No. CF6–80C2 S/B 72–A1024, Revision 1, dated November 3, 2000, or any version of GE ASB No. CF6–80C2 S/B 72–A1026, inspect, rework, re-mark, and remove from service as necessary using the compliance times in the following Table 7:

**TABLE 7.—COMPLIANCE TIMES FOR INSPECTION AND REWORK OF CF6–80C2 SERIES STAGE 1 HPT ROTOR DISKS, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, AND 1531M84G10 WITH SNS NOT LISTED IN TABLE 6 AND TABLE 6A OF THIS AD—PREVIOUSLY INSPECTED**

Stage 1 HPT rotor disk cycles-since-last-inspection (CSLI) on March 12, 2004 (effective date of superseded AD 2004–04–07)	Compliance time for inspection and rework
(i) More than 1,500 CSLI .....	At the next ESV after March 12, 2004 (effective date of superseded AD 2004–04–07), but not to exceed 4,500 CSLI.
(ii) 1,500 CSLI or fewer .....	At the next ESV after March 12, 2004 (effective date of superseded AD 2004–04–07), but not to exceed 3,500 CSLI.

(3) For stage 1 HPT rotor disks that have not been inspected before March 12, 2004 (effective date of superseded AD 2004–04–07) using GE ASB No. CF6–80C2 S/B 72–

A1024, Revision 1, dated November 3, 2000, or any version of GE ASB No. CF6–80C2 S/B 72–A1026, inspect, rework, re-mark, and remove from service as necessary using the

following Table 8 or Table 8A compliance times, whichever occurs first:

**TABLE 8.—COMPLIANCE TIMES FOR INSPECTION AND REWORK OF CF6–80C2 SERIES STAGE 1 HPT ROTOR DISKS, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, AND 1531M84G10 WITH SNS NOT LISTED IN TABLE 6 AND TABLE 6A OF THIS AD—NOT PREVIOUSLY INSPECTED**

Stage 1 HPT rotor disk cycles-since-new (CSN) on the effective date of this AD	Compliance time for inspection and rework
(i) 9,000 or more CSN .....	Within 250 CIS after the effective date of this AD, or by March 31, 2007, whichever occurs first.
(ii) 6,900 or more but fewer than 9,000 CSN .....	Within 500 CIS after the effective date of this AD, but before accumulating 9,250 CSN, or by December 31, 2007, whichever occurs first.
(iii) Fewer than 6,900 CSN .....	Before accumulating 7,400 CSN, or by December 31, 2008, whichever occurs first.

**TABLE 8A.—COMPLIANCE TIMES FOR INSPECTION AND REWORK OF CF6–80C2 SERIES STAGE 1 HPT ROTOR DISKS, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, AND 1531M84G10 WITH SNS NOT LISTED IN TABLE 6 AND TABLE 6A OF THIS AD—NOT PREVIOUSLY INSPECTED**

Stage 1 HPT rotor disk CSN on March 12, 2004 (effective date of superseded AD 2004–04–07)	Compliance time for inspection and rework
(i) 10,000 or more CSN .....	At the next ESV or within 1,000 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first.
(ii) 5,000 or more CSN but fewer than 10,000 CSN .....	At the next ESV or within 2,400 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), but before accumulating 11,000 CSN.
(iii) Fewer than 5,000 CSN .....	At the next ESV or within 3,500 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first, but before accumulating 7,400 CSN.

Stage 1 HPT Rotor Disks, P/N 1862M23G01

(m) For stage 1 HPT rotor disk, P/N 1862M23G01, remove the disk from service at the following times:

(1) For stage 1 HPT rotor disks not installed in engines, remove from service as necessary before further flight.

(2) For stage 1 HPT rotor disks that have been inspected before March 12, 2004 (effective date of superseded AD 2004–04–07), using any version of GE ASB No. CF6–80C2 S/B 72–A1026, and had more than zero CSN at the time of that inspection, remove from service at next ESV.

(3) For stage 1 HPT rotor disks that have not been inspected, or were only inspected with zero CSN before March 12, 2004 (effective date of superseded AD 2004–04–07), using any version of GE ASB No. CF6–80C2 S/B 72–A1026, remove from service

using the following Table 9 or Table 9A compliance times, whichever occurs first:

TABLE 9.—COMPLIANCE TIMES FOR REMOVAL OF CF6–80C2 SERIES STAGE 1 HPT ROTOR DISKS, P/N 1862M23G01—NOT PREVIOUSLY INSPECTED

Stage 1 HPT rotor disk CSN on the effective date of this AD	Compliance time for removal
(i) 9,000 or more CSN .....	Within 250 CIS after the effective date of this AD, or by March 31, 2007, whichever occurs first.
(ii) 6,900 or more but fewer than 9,000 CSN .....	Within 500 CIS after the effective date of this AD, but before accumulating 9,250 CSN, or by December 31, 2007, whichever occurs first.
(iii) Fewer than 6,900 CSN .....	Before accumulating 7,400 CSN, or by December 31, 2008, whichever occurs first.

TABLE 9A.—COMPLIANCE TIMES FOR REMOVAL OF CF6–80C2 SERIES STAGE 1 HPT ROTOR DISKS, P/N 1862M23G01—NOT PREVIOUSLY INSPECTED

Stage 1 HPT rotor disk CSN on March 12, 2004 (effective date of superseded AD 2004–04–07)	Compliance time for removal
(i) 10,000 or more CSN .....	At the next ESV or within 1,000 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first.
(ii) 5,000 or more CSN but fewer than 10,000 CSN .....	At the next ESV or within 2,400 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first, but before accumulating 11,000 CSN.
(iii) Fewer than 5,000 CSN .....	At the next ESV or within 3,500 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first, but before accumulating 7,400 CSN.

CF6–80E1A2, A4 Engines

Stage 1 HPT Rotor Disks, P/N 1639M41P04

(n) For stage 1 HPT rotor disks, P/N 1639M41P04, remove the rotor disks from

service using paragraphs 3.A.(1) through 3.A.(2) of Accomplishment Instructions of GE SB No. CF6–80E1 S/B 72–0251, dated January 22, 2004, at the following times:

(1) For stage 1 HPT rotor disks currently in service, remove the disk using the compliance times in the following Table 10 or Table 10A compliance times, whichever occurs first:

TABLE 10.—COMPLIANCE TIMES FOR REMOVAL OF CF6–80E1 STAGE 1 HPT ROTOR DISKS, P/N 1639M41P04

Stage 1 HPT rotor disk CSN on the effective date of this AD	Compliance Time For Removal
(i) 9,000 or more CSN .....	Within 250 CIS after the effective date of this AD, or by March 31, 2007, whichever occurs first.
(ii) 6,900 or more but fewer than 9,000 CSN .....	Within 500 CIS after the effective date of this AD, but before accumulating 9,250 CSN, or by December 31, 2007, whichever occurs first.
(iii) Fewer than 6,900 CSN .....	Before accumulating 7,400 CSN, or by December 31, 2008, whichever occurs first.

TABLE 10A.—COMPLIANCE TIMES FOR REMOVAL OF CF6–80E1 STAGE 1 HPT ROTOR DISKS, P/N 1639M41P04

Stage 1 HPT rotor disk CSN on the March 12, 2004 ( effective date of superseded AD 2004–04–07)	Compliance time for removal
(i) More than 10,000 CSN .....	At the next ESV or within 600 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first.
(ii) More than 5,000 CSN but fewer than or equal to 10,000 CSN .....	At the next ESV or within 2,500 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first, but before accumulating 10,600 CSN.
(iii) Fewer than or equal to 5,000 CSN .....	At the next ESV or within 3,500 CIS after March 12, 2004 (effective date of superseded AD 2004–04–07), whichever occurs first, but before accumulating 7,500 CSN.

(2) After March 12, 2004 (effective date of superseded AD 2004–04–07), do not install any stage 1 HPT rotor disk, P/N 1639M41P04, into any engine.

**Definitions**

(o) For the purpose of this AD, the following definitions apply:

(1) An engine shop visit (ESV) is when the engine is removed from an aircraft for maintenance and a major engine flange is disassembled. For stage 1 HPT rotor disks that have been inspected using any version of GE SB No. CF6–80A SB 72–0779 or any version of GE ASB No. CF6–80C2 ASB 72–A1026 or GE SB No. CF6–80C2 SB 72–A1024, Revision 1, dated November 3, 2000 or are

listed in Table 6A or Table 6B, the following actions, either separately or in combination with each other, are not considered ESVs for the purpose of this AD:

- (i) The removal of the upper compressor stator case solely for airfoil maintenance.
- (ii) The module level inspection of the high-pressure compressor rotor 3–9 spool.



(iii) The replacement of stage 5 high-pressure compressor variable stator vane bushings or lever arms.

(2) Piece-part exposure is when according to the manufacturer's engine manual or other FAA-approved engine manual the stage 1 HPT rotor disk is considered completely disassembled.

**Reporting Requirements**

(p) Within five calendar days of the inspection, report the results of inspections that equal or exceed the reject criteria to: Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803; telephone (781) 238-7176; fax (781) 238-7199. Reporting requirements have been approved by the Office of Management and Budget and assigned OMB control number 2120-0056. Be sure to include the following information:

- (1) Engine model in which the stage 1 HPT rotor disk was installed.
- (2) Part Number.
- (3) Serial Number.

- (4) Part CSN.
- (5) Part CSLI.

(6) Date and location where inspection was done.

(g) We request that you record the inspection information and results on GE Form 1653-1, entitled CF6-80A/80C Stage 1 HPT Disk Dovetail Slot Bottom Inspection. This form is available in any version of GE SB CF6-80A S/B 72-0779, or GE ASB CF6-80C2 S/B 72-A1026. We also request that a copy of the data be sent to GE Airline Support Engineering, General Electric Aircraft Engines, Customer Support Center, 1 Neumann Way, Mail Drop RM285, Cincinnati, OH 45215.

**Alternative Methods of Compliance**

(r) The manager, Engine Certification Office, has the authority to approve alternative methods of compliance for this AD if requested using the procedures found in 14 CFR 39.19.

**Material Incorporated by Reference**

(s) You must use the service information specified in Table 11 to perform the actions

required by this AD. The Director of the Federal Register previously approved the incorporation by reference of General Electric Service Bulletins No. CF6-80E1 S/B 72-0251, dated January 22, 2004 and No. CF6-80A S/B 72-0779, Revision 1, dated January 22, 2004, and Alert Service Bulletin No. CF6-80C2 S/B 72-A1026, Revision 2, dated January 22, 2004, as of March 12, 2004 (69 FR 8801, February 26, 2004). The Director of the Federal Register approved the incorporation by reference of the other documents listed in Table 11 of this AD in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. You can get a copy from General Electric Company via Lockheed Martin Technology Services, 10525 Chester Road, Suite C, Cincinnati, Ohio 45215, telephone (513) 672-8400, fax (513) 672-8422. You may review copies at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA; or at the Office of the Federal Register, 800 North Capitol Street, NW, Suite 700, Washington, DC. Table 11 follows:

TABLE 11.—INCORPORATION BY REFERENCE

Service Bulletin No.	Page	Revision	Date
GE SB No. CF6-80E1 S/B 72-0251 Total Pages: 4	All	Original	January 22, 2004.
GE SB No. CF6-80A S/B 72-0779 Total Pages: 34	ALL	1	January 22, 2004.
GE SB No. CF6-80A S/B 72-0788 Total Pages: 11	ALL	3	July 20, 2006.
GE ASB No. CF6-80C2 S/B 72-A1026 Total Pages: 38	ALL	2	January 22, 2004.
GE SB No. CF6-80C2 S/B 72-1089 Total Pages: 11	ALL	3	July 20, 2006.
GE SB No. CF6-80C2 S/B 72-1217 Total Pages: 12	ALL	Original	July 20, 2006.
GE SB No. CF6-80A S/B 72-0822 Total Pages: 10	ALL	Original	July 20, 2006.

**Related Information**

(t) GE ASB No. CF6-80C2 S/B 72-A1024, Revision 1, dated November 3, 2000 also pertains to the subject of this AD.

Issued in Burlington, Massachusetts, on August 10, 2006.

**Francis A. Favara,**

*Manager, Engine and Propeller Directorate, Aircraft Certification Service.*

[FR Doc. E6-13437 Filed 8-17-06; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION  
Federal Aviation Administration**

**14 CFR Part 39**

[Docket No. FAA-2006-24366; Directorate Identifier 2006-NM-040-AD; Amendment 39-14716; AD 2006-16-16]

**RIN 2120-AA64**

**Airworthiness Directives; Empresa Brasileira de Aeronautica S.A. (EMBRAER) Model EMB-135BJ Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** The FAA is adopting a new airworthiness directive (AD) for certain EMBRAER Model EMB-135BJ airplanes. This AD requires inspecting for missing fire blocking material on the left- and

right-hand partitions of the forward baggage compartment door; replacing the seal on both partitions; and performing corrective action if necessary. This AD results from a report indicating that certain airplanes were delivered with the fire blocking material missing and the seal improperly installed on the partitions of the forward baggage compartment door. We are issuing this AD to detect and correct such discrepancies on the forward baggage compartment partition, which, in the event of a fire in the baggage compartment, could result in smoke propagating into the main cabin.

**DATES:** This AD becomes effective September 22, 2006.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in the AD as of September 22, 2006.

**ADDRESSES:** You may examine the AD docket on the Internet at <http://dms.dot.gov> or in person at the Docket