

head office of the Federal Reserve Bank of Cleveland. The depository institutions that are located in the affected check processing regions and that include the routing symbols in their disclosure statements would be required to notify customers of the resulting change in availability under § 229.18(e). However, because all paperwork collection procedures associated with Regulation CC already are in place, the Board anticipates that no additional burden will be imposed as a result of this rulemaking.

**List of Subjects in 12 CFR Part 229**

Banks, Banking, Federal Reserve System, Reporting and recordkeeping requirements.

**Authority and Issuance**

■ For the reasons set forth in the preamble, the Board is amending 12 CFR part 229 to read as follows:

**PART 229—AVAILABILITY OF FUNDS AND COLLECTION OF CHECKS (REGULATION CC)**

■ 1. The authority citation for part 229 continues to read as follows:

*Authority:* 12 U.S.C. 4001 *et seq.*

■ 2. The first sentence of paragraph A and the Fourth Federal Reserve District routing symbol list in appendix A are revised to read as follows:

**Appendix A to Part 229—Routing Number Guide to Next-Day Availability Checks and Local Checks**

A. Each bank is assigned a routing number by an agent of the American Bankers Association. \* \* \*

\* \* \* \* \*

**Fourth Federal Reserve District**

[Federal Reserve Bank of Cleveland]

*Head Office*

- 0410 2410
- 0412 2412
- 0430 2430
- 0432 2432
- 0433 2433
- 0434 2434

*Cincinnati Branch*

- 0420 2420
- 0421 2421
- 0422 2422
- 0423 2423

*Columbus Office*

- 0440 2440
- 0441 2441
- 0442 2442

\* \* \* \* \*

■ 3. Appendix E is amended by removing the phrase “Thomson Financial Publishing Inc.” in sections II.DD., XVIII.A.2.b.ii., and XXII.B.2.b.i.

and adding the phrase “an agent of the American Bankers Association” in its place.

By order of the Board of Governors of the Federal Reserve System, August 27, 2003.

**Jennifer J. Johnson,**

*Secretary of the Board.*

[FR Doc. 03-22333 Filed 8-29-03; 8:45 am]

**BILLING CODE 6210-01-P**

**DEPARTMENT OF TRANSPORTATION**

**Federal Aviation Administration**

**14 CFR Part 39**

[Docket No. 2000-CE-64-AD; Amendment 39-13291; AD 2003-17-16]

RIN 2120-AA64

**Airworthiness Directives; Robert E. Rust Models DeHavilland DH.C1 Chipmunk 21, 22, and 22A Airplanes**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD) that applies to certain Robert E. Rust (R.E. Rust) Models DeHavilland DH.C1 Chipmunk 21, 22, and 22A airplanes. This AD requires you to repetitively inspect the tailplane attachment brackets and replace each bracket at a specified time. This AD also requires you to repetitively inspect each joint of the port and starboard engine mount frame and the rear upper mount frame tubes for cracks and/or damage and repair any cracks and/or damage found. This AD is the result of reports of stress corrosion cracking found on the tailplane attachment brackets and fatigue cracking and chaffing of the engine mount frame. The actions specified by this AD are intended to prevent failure of the tailplane attachment brackets caused by stress corrosion cracking and failure of the engine mount, which could result in loss of the tail section and separation of the engine from the airplane respectively. Such failures could lead to loss of control of the airplane.

**DATES:** This AD becomes effective on October 10, 2003.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in the regulations as of October 10, 2003.

**ADDRESSES:** You may get the service information referenced in this AD from DeHavilland Support Limited, Duxford Airfield, Bldg. 213, Cambridgeshire, CB2 4QR, United Kingdom, telephone:

+44 1223 830090, facsimile: +44 1223 830085, e-mail: *info@dhsupport.com*. You may view this information at the Federal Aviation Administration (FAA), Central Region, Office of the Regional Counsel, Attention: Rules Docket No. 2000-CE-64-AD, 901 Locust, Room 506, Kansas City, Missouri 64106; or at the Office of the Federal Register, 800 North Capitol Street NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Cindy Lorenzen, Aerospace Engineer, FAA, Atlanta Aircraft Certification Office, 1895 Phoenix Boulevard, suite 450, Atlanta, Georgia; telephone: (770) 703-6078; facsimile: (770) 703-6097.

**SUPPLEMENTARY INFORMATION:**

**Discussion**

*What Events Have Caused This AD?*

We received reports that an unsafe condition exists on certain R.E. Rust Models DeHavilland DH.C1 Chipmunk 21, 22, and 22A airplanes. After reviewing several of these airplanes, stress corrosion cracking was found on the tailplane attachment brackets and fatigue cracks and chaffing were found on the engine mount frame.

Cracks in the engine mount frame were found in the area of the junction of the front and rear top tube and engine mounting foot support brackets and in the front of the frame. We have determined that fatigue is the cause of the cracks. The upper aft mount frame tubes were also found to have damage caused by chaffing by the cowling support rod.

*What Is the Potential Impact if FAA Took No Action?*

These conditions, if not corrected, could result in failure of the tailplane attachment brackets and failure of the engine mount. Such failures could lead to loss of control of the airplane.

*Has FAA Taken Any Action to This Point?*

We issued a proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an AD that would apply to certain R.E. Rust Models DeHavilland DH.C1 Chipmunk 21, 22, and 22A airplanes. This proposal was published in the **Federal Register** as a supplemental notice of proposed rulemaking (NPRM) on April 15, 2003 (68 FR 18571). The supplemental NPRM proposed to require you to repetitively inspect the tailplane attachment brackets and replace each bracket at a specified time. The NPRM also proposed to require you to repetitively inspect each joint of the port and starboard engine mount frame and the

rear upper mount frame tubes for cracks and/or damage and repair any cracks and/or damage found.

*Was the Public Invited To Comment?*

The FAA encouraged interested persons to participate in the making of this amendment. We did not receive any comments on the proposed rule or on our determination of the cost to the public.

**FAA's Determination**

*What Is FAA's Final Determination on This Issue?*

After careful review of all available information related to the subject presented above, we have determined that air safety and the public interest

require the adoption of the rule as proposed except for minor editorial corrections. We have determined that these minor corrections:

- Provide the intent that was proposed in the supplemental NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the Supplemental NPRM.

*How Does the Revision to 14 CFR Part 39 Affect This AD?*

On July 10, 2002, FAA published a new version of 14 CFR part 39 (67 FR 47997, July 22, 2002), which governs FAA's AD system. This regulation now includes material that relates to special flight permits, alternative methods of

compliance, and altered products. This material previously was included in each individual AD. Since this material is included in 14 CFR part 39, we will not include it in future AD actions.

**Cost Impact**

*How Many Airplanes Does This AD Impact?*

We estimate that this AD affects 54 airplanes in the U.S. registry.

*What Is the Cost Impact of This AD on Owners/Operators of the Affected Airplanes?*

We estimate the following costs to accomplish an inspection of the tailplane attachment brackets:

Labor cost	Parts cost	Total cost per airplane	Total cost on U.S. operators
32 workhours × \$60 per hour = \$1,920 .....	No parts required .....	\$1,920	\$1,920 × 54 = \$103,680

We estimate the following costs to accomplish any necessary replacements that would be required based on the

results of the proposed inspection. We have no way of determining the number

of airplanes that may need such replacement:

Labor cost	Parts cost	Total cost per bracket
3 workhours × \$60 per hour = \$180 per bracket .....	\$600 per bracket (2 brackets per airplane) .....	\$180 + 600 = \$780

We estimate the following costs to accomplish an inspection of the engine mount frame:

Labor cost	Parts cost	Total cost per airplane	Total cost on U.S. operators
16 workhours × \$60 per hour = \$960 .....	No parts required .....	\$960	\$960 × 54 = \$51,840

The FAA has no method of determining the number of repairs or replacements each owner/operator will incur over the life of each of the affected airplanes based on the results of the proposed inspections. We have no way of determining the number of airplanes that may need such repair. The extent of damage may vary of each airplane.

**Compliance Time of This AD**

*What Will Be the Compliance Time of This AD?*

The compliance time for the initial inspections in this AD is "within the next 90 days after the effective date of this AD."

*Why Is the Compliance Time Presented in Calendar Time Instead of Hours Time-in-Service (TIS)?*

An unsafe condition specified by this AD is caused by corrosion. Corrosion can occur regardless of whether the aircraft is in operation or is in storage. Therefore, to assure that the unsafe condition specified in this AD does not go undetected for a long period of time, the compliance is presented in calendar time instead of TIS.

**Regulatory Impact**

*Does This AD Impact Various Entities?*

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various

levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

*Does This AD Involve a Significant Rule or Regulatory Action?*

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the final evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the

Rules Docket at the location provided under the caption **ADDRESSES**.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**Adoption of the Amendment**

Accordingly, under the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

2. FAA amends § 39.13 by adding a new AD to read as follows:

**2003-17-16 Robert E. Rust:** Amendment 39-13291; Docket No. 2000-CE-64-AD.

(a) *What airplanes are affected by this AD?*

This AD affects R.E. Rust Models DeHavilland DH.C1 Chipmunk 21, 22, and 22A airplanes, serial numbers C1-001 through C1-1014, that are type certificated in any category.

**Note 1:** We recommend all owners/operators of DeHavilland DH.C1 Chipmunk 21, 22, and 22A airplanes, serial numbers C1-001 through C1-1014, with experimental airworthiness certificates comply with the actions required in this AD.

(b) *Who must comply with this AD?*

Anyone who wishes to operate any of the airplanes identified in paragraph (a) of this AD must comply with this AD.

(c) *What problem does this AD address?*

The actions specified by this AD are intended to prevent failure of the tailplane attachment brackets caused by stress corrosion cracking and failure of the engine mount, which could result in loss of the tail section and separation of the engine from the airplane respectively. Such failures could lead to loss of control of the airplane.

(d) *What actions must I accomplish to address this problem?*

To address this problem, you must accomplish the following:

**(1) TAILPLANE ATTACHMENT BRACKETS**

Compliance	Actions	Procedures
(i) Initially inspect within the next 90 days after October 10, 2003 (the date of this AD). (A) Inspect thereafter at intervals not to exceed 6 months or 150 fatigue hours, whichever occurs first, until the modification required by paragraph (d)(1)(ii) of this AD is incorporated. (B) When the modification required by paragraph (d)(1)(ii) is incorporated, you may terminate the repetitive inspections of the tailplane attachment brackets.	Inspect, using dye penetrant, the tailplane attachment brackets, part-number (P/N) C1.TP.167 (or FAA-approved equivalent part) for cracks.	In accordance with British Aerospace Military Aircraft and Aerostructures (BAe Aircraft) Mandatory Technical News Sheet CT (C1) No. 176, Issue 2, dated November 1, 1997; and Civil Modification Mandatory Modification No. Chipmunk H357, dated March 12, 1984. Calculate fatigue hours by multiplying the TIS by the role factor in accordance with British Aerospace Mandatory Technical News Sheet Series: Chipmunk (C1), No. 138, Issue: 5, dated August 1, 1985.
(ii) At whichever of the following that occurs first: (A) Prior to further flight after the inspection where any crack is found; or (B) Upon accumulating 9,984 fatigue hours or within the next 90 days after October 10, 2003 (the effective date of this AD), whichever occurs later.	Replace the tailplane attachment bracket by incorporating Modification H357 (P/N C1.TP.313) or FAA-approved equivalent part number. Installing P/N C1.TP.313 (or FAA-approved equivalent part number) terminates the repetitive inspection requirement of the tailplane attachment brackets.	In accordance with British Aerospace Military Aircraft and Aerostructures (BAe Aircraft) Mandatory Technical News Sheet CT (C1) No. 176, Issue 2, dated November 1, 1997; and Civil Modification No. Chipmunk H357, dated March 12, 1984. Calculate fatigue hours by multiplying the TIS by the role in accordance with British Aerospace Mandatory Technical News Sheet Series: Chipmunk (C1), No. 138, Issue: 5, dated August 1, 1985.
(iii) As of October 10, 2003 (the effective date of this AD).	Only install a tailplane attachment bracket that is P/N C1.TP.313. or FAA-approved equivalent part number.	Not applicable.
(iv) As of October 10, 2003 (the effective date of this AD).	Incorporate the following into the Aircraft Logbook: "In accordance with AD 2003-17-16, the tailplane attachment bracket is life limited to 9,984 fatigue hours."	In accordance with British Aerospace Military Aircraft and Aerostructures (BAe Aircraft) Mandatory Technical News Sheet CT (C1) No. 176, Issue 2, dated November 1, 1997.

**(2) ENGINE MOUNT FRAMES**

Actions	Compliance	Procedures
(i) Inspect each joint of the port and starboard engine mount and the rear upper mount frame tubes for cracks and damage.	Initially inspect within the next 90 days after October 10, 2003 (the effective date of this AD). Repetitively inspect thereafter at intervals not to exceed 600 hours TIS.	In accordance with British Aerospace Aerostructures Limited (BAe Aircraft) Mandatory Technical News Sheet CT (C1) No. 190, Issue 2, dated April 1, 1995.

## (2) ENGINE MOUNT FRAMES—Continued

Actions	Compliance	Procedures
<p>(ii) If cracks or damage is found during any inspection required in paragraph (d)(2)(i) of this AD:</p> <p>(A) obtain a repair scheme from the manufacturer through the FAA at the address specified in paragraph (e) of this AD and incorporate this repair scheme, or repair in accordance with FAA Advisory Circular (AC) 43.13-1B, Change 1, dated September 27, 2001, Chapter 4, Paragraph 4-99; or</p> <p>(B) replace with a new or serviceable part.</p>	<p>Prior to further flight after the inspection in which any crack or damage is found. Repeatedly inspect as required in paragraph (d)(2)(i) of this AD.</p>	<p>Repair in accordance with AC 43.13-1B, Change 1, dated September 27, 2001, Chapter 4, Paragraph 4-99 or in accordance with the repair scheme obtained from DeHavilland Support Limited, Duxford Airfield, Bldg. 213, Cambridgeshire, CB2 4QR, United Kingdom. Obtain this repair scheme through the FAA at the address specified in paragraph (f) of this AD. Replace in accordance with British Aerospace Aerostructures Limited (BAe Aircraft) Mandatory Technical News Sheet CT (C1) No. 190, Issue 2, dated April 1, 1995, or AC 43.13-1B, Change 1, dated September 27, 2001, Chapter 4, Paragraph 4-99.</p>
<p>(iii) Bind the rear upper mount frame tubes with a high density polythene tape at the location where the cowling support rod clip is secured.</p>	<p>Prior to further flight after any inspection required in paragraph (d)(2)(i) of this AD if no cracks or damage is found, and prior to further flight after any repairs or replacement is made as required in paragraph (d)(2)(ii) of this AD .</p>	<p>In accordance with British Aerospace Aerostructures Limited (BAe Aircraft) Mandatory Technical News Sheet CT (C1) No. 190, Issue 2, dated April 1, 1995.</p>

(e) *Can I comply with this AD in any other way?* To use an alternative method of compliance or adjust the compliance time, use the procedures in 14 CFR 39.19. Send these requests to the Manager, Atlanta Aircraft Certification Office (ACO). Contact Cindy Lorenzen, Aerospace Engineer, FAA, Atlanta Aircraft Certification Office, 1895 Phoenix Boulevard, Suite 450, Atlanta, Georgia; telephone: (770) 703-6078; facsimile: (770) 703-6097.

(f) *Are any service bulletins incorporated into this AD by reference?* Actions required by this AD must be done in accordance with British Aerospace Military Aircraft and Aerostructures (BAe Aircraft) Mandatory Technical News Sheet CT (C1) No. 176, Issue 2, dated November 1, 1997; Civil Modification Mandatory Modification No. Chipmunk H357, dated March 12, 1984; British Aerospace Mandatory Technical News Sheet Series: Chipmunk (C1), No. 138, Issue: 5, dated August 1, 1985; and British Aerospace Aerostructures Limited (BAe Aircraft) Mandatory Technical News Sheet CT (C1) No. 190, Issue 2, including Appendix 1 (Part B), dated April 1, 1995. The Director of the Federal Register approved this incorporation by reference under 5 U.S.C. 552(a) and 1 CFR part 51. You may get copies from DeHavilland Support Limited, Duxford Airfield, Bldg. 213, Cambridgeshire, CB2 4QR, United Kingdom, telephone: +44 1223 830090, facsimile: +44 1223 830085, e-mail: [info@dhsupport.com](mailto:info@dhsupport.com). You may view copies at the FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri, or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(g) *When does this amendment become effective?* This amendment becomes effective on October 10, 2003.

Issued in Kansas City, Missouri, on August 19, 2003.

**Michael Gallagher,**

*Manager, Small Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 03-21742 Filed 8-29-03; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2002-NM-74-AD; Amendment 39-13287; AD 2003-17-12]

**RIN 2120-AA64**

#### **Airworthiness Directives; McDonnell Douglas Model MD-11 and MD-11F Airplanes**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to certain McDonnell Douglas Model MD-11 and MD-11F airplanes, that requires a one-time visual inspection of the circuit breakers to determine if discrepant circuit breakers are installed, and corrective action if necessary. This action is necessary to prevent internal overheating and arcing of circuit breakers and airplane wiring due to long-term use and breakdown of internal components of the circuit breakers, which could result in smoke and fire in the flight compartment and

main cabin. This action is intended to address the identified unsafe condition.

**DATES:** Effective October 7, 2003.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of October 7, 2003.

**ADDRESSES:** The service information referenced in the proposed rule may be obtained from Boeing Commercial Aircraft Group, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1-L5A (D800-0024). This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California; or at the Office of the Federal Register, 800 North Capitol Street NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Natalie Phan-Tran, Aerospace Engineer, Systems and Equipment Branch, ANM-130L, FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California 90712-4137; telephone (562) 627-5343; fax (562) 627-5210.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain McDonnell Douglas Model MD-11 and MD-11E airplanes series airplanes was published in the **Federal Register** on June 11, 2003 (68 FR 34847). That action proposed to