

bulletin is intended to adequately address the identified unsafe condition.

Explanation of Requirements of Proposed Rule

Since an unsafe condition has been identified that is likely to exist or develop on other products of this same type design, the proposed AD would require accomplishment of the actions specified in the service bulletin described previously, except as discussed below.

Differences Between Proposed Rule and Service Bulletin

Operators should note that, although the service bulletin recommends accomplishing the replacement at the next normally scheduled maintenance period, the FAA has determined that such an imprecise compliance time would not address the identified unsafe condition in a timely manner. In developing an appropriate compliance time for this proposed AD, the FAA considered not only the manufacturer's recommendation, but the degree of urgency associated with addressing the subject unsafe condition, the average utilization of the affected fleet, and the time necessary to perform the inspection (three hours). In light of all of these factors, the FAA finds an 18-month compliance time for completing the required actions to be warranted, in that it represents an appropriate interval of time allowable for affected airplanes to continue to operate without compromising safety.

Cost Impact

There are approximately 360 airplanes of the affected design in the worldwide fleet. The FAA estimates that 124 airplanes of U.S. registry would be affected by this proposed AD, that it would take approximately 3 work hours per airplane to accomplish the proposed replacement, and that the average labor rate is \$65 per work hour. Required parts would cost approximately \$2,200 per airplane. Based on these figures, the cost impact of the proposed AD on U.S. operators is estimated to be \$296,980, or \$2,395 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this proposed AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time

required to gain access and close up, planning time, or time necessitated by other administrative actions. The manufacturer may cover the cost of replacement parts associated with this proposed AD, subject to warranty conditions. Manufacturer warranty remedies may also be available for labor costs associated with this proposed AD. As a result, the costs attributable to the proposed AD may be less than stated above.

Regulatory Impact

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this proposal would not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption ADDRESSES.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Boeing: Docket 2002–NM–14–AD.

Applicability: Model 777 series airplanes, as listed in Boeing Service Bulletin 777–25–

0191, dated September 13, 2002, certificated in any category.

Compliance: Required as indicated, unless accomplished previously.

To prevent uncommanded movements of the power drive unit during ground handling of cargo and consequent possible injury to ground personnel, accomplish the following:

Replacement

(a) Within 18 months after the effective date of this AD, replace the cargo control joysticks with new joysticks, per the Accomplishment Instructions of Boeing Service Bulletin 777–25–0191, dated September 13, 2002.

Parts Installation

(b) As of the effective date of this AD, no person shall install a cargo control joystick, part number S283W602–1 or S283W602–2, on any airplane.

Alternative Methods of Compliance

(c) In accordance with 14 CFR 39.19, the Manager, Seattle Aircraft Certification Office, FAA, is authorized to approve alternative methods of compliance for this AD.

Issued in Renton, Washington, on November 21, 2003.

Vi L. Lipski,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03–29697 Filed 11–26–03; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2003–NM–154–AD]

RIN 2120–AA64

Airworthiness Directives; Bombardier Model DHC–8–102, –103, –106, –201, –202, –301, –311, and –315 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain Bombardier Model DHC–8–102, –103, –106, –201, –202, –301, –311, and –315 series airplanes. This proposal would require repetitive inspections for discrepancies of certain rear spar fittings between the flex shaft of the flap secondary drive and the wing-to-fuselage structure, and corrective action if necessary. This proposal also provides for an optional modification of the flex shaft installation, which would terminate the repetitive inspections. This action is necessary to find and fix

damage and subsequent failure of the rear spar fittings, which could result in loss of the wing. This action is intended to address the identified unsafe condition.

DATES: Comments must be received by December 29, 2003.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2003-NM-154-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227-1232. Comments may also be sent via the Internet using the following address: *9-anm-nprmcomment@faa.gov*. Comments sent via fax or the Internet must contain "Docket No. 2003-NM-154-AD" in the subject line and need not be submitted in triplicate. Comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 or 2000 or ASCII text.

The service information referenced in the proposed rule may be obtained from Bombardier, Inc., Bombardier Regional Aircraft Division, 123 Garratt Boulevard, Downsview, Ontario M3K 1Y5, Canada. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York.

FOR FURTHER INFORMATION CONTACT: Jon Hjelm, Aerospace Engineer, Airframe and Propulsion Branch, ANE-171, FAA, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York 11581; telephone (516) 256-7523; fax (516) 568-2716.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this action may be changed in light of the comments received.

Submit comments using the following format:

- Organize comments issue-by-issue. For example, discuss a request to change the compliance time and a request to change the service bulletin reference as two separate issues.

- For each issue, state what specific change to the proposed AD is being requested.

- Include justification (*e.g.*, reasons or data) for each request.

- Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this action must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 2003-NM-154-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2003-NM-154-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

Discussion

Transport Canada Civil Aviation (TCCA), which is the airworthiness authority for Canada, notified the FAA that an unsafe condition may exist on certain Bombardier Model DHC-8 series airplanes. TCCA has informed the FAA that discrepancies (chafing, wear damage, cracking) have been found on the rear spar fittings (part numbers (P/N) 85320053, 85322060, and 85334180), located between the flex shaft of the flap secondary drive and the wing-to-fuselage structure. These discrepancies are due to inadequate clearance between the fittings and the flex shaft of the flap secondary drive mechanism, caused by vibration of the flex drive during flap extension/retraction. Such discrepancies could affect the fatigue life of the fittings, which could result in failure of the fittings and consequent loss of the wing.

Explanation of Relevant Service Information

Bombardier has issued Service Bulletin 8-27-83, Revision 'A', dated February 8, 2002, which describes

procedures for repetitive inspections for discrepancies (chafing, wear damage, cracking) of certain rear spar fittings between the flex shaft of the flap secondary drive and the wing-to-fuselage structure, and corrective action if necessary. The service bulletin also provides procedures for an optional modification of the flex shaft, which would eliminate the need for the repetitive inspections. The inspections and corrective action are as follows:

- A visual inspection to determine the wear damage of each rear spar fitting, which includes the following actions:

If wear damage is found, measure the depth of the wear; and if wear depth is less than the limits specified in Table 1 of the service bulletin, continued operation is allowed for 4,000 flight cycles without blending out the wear; when 4,000 flight cycles have been accumulated, the wear damage must be blended out and must be within the limits specified in Table 3 of the service bulletin. After blending the fitting must be re-inspected (high frequency eddy current (HFEC) inspection) for any remaining discrepancies (wear, cracking). Discrepancies must be repaired before further flight. If no discrepancies are found the inspection is to be repeated at intervals not to exceed 12 months.

If the wear depth is outside the limits specified in Table 1 of the service bulletin, but is less than the limits specified in Table 2 of the service bulletin, temporary operation is allowed for 400 flight cycles without blending out the wear; when 400 flight cycles have been accumulated, the wear must be blended out and within the limits specified in Table 3. The inspection is to be repeated at intervals not to exceed 12 months.

If the wear depth is greater than the limits specified in Table 2, or after blending is greater than the limits specified in Table 3, or cracking is found after temporary operation, the fitting must be replaced before further flight.

- A HFEC inspection for cracking of damaged areas after continued operation and after blending out wear damage. If no cracking is found and the blended wear is within the limits specified in Table 3, permanent continued operation is allowed. If cracking is found or blended wear exceeds the limits specified in Table 3, the fitting must be replaced before further flight.

- Replacement of the rear spar fittings includes removal of the existing fittings, removal of old sealant, inspection of each hole through the rear spar and fuselage for damage, repair of any

damage before further flight, and application of new sealant, installation of new fittings, and application of anti-corrosive compound.

- The optional modification of the flex shaft includes installation of new brackets on the rear spar, rework of the torque tube support fittings in the flap primary drive, installation of a new torque tube retainer tray assembly, and installation of additional clamps to stabilize the flex shaft.

The service bulletin also describes procedures for functional tests after doing all applicable actions.

Accomplishment of the actions specified in the service bulletin is intended to adequately address the identified unsafe condition. TCCA classified this service bulletin as mandatory and issued Canadian airworthiness directive CF-2001-42, dated November 23, 2001, to ensure the continued airworthiness of these airplanes in Canada.

FAA's Conclusions

These airplane models are manufactured in Canada and are type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, TCCA has kept us informed of the situation described above. We have examined the findings of TCCA, reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

Explanation of Requirements of Proposed AD

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design registered in the United States, the proposed AD would require accomplishment of the actions specified in the service bulletin described previously, except as discussed below.

This AD allows flight with wear damage, provided that (1) the wear damage is within the limits specified in the service bulletin, (2) no cracking is found, and (3) established inspection procedures would find wear damage in structure at intervals permitting repairs to be done before reduced structural integrity of the fuselage could occur.

To be consistent with the findings of the TCCA, this proposed AD allows operators to continue the repetitive inspections instead of doing the terminating action. In making this determination, we consider that, in the

case of this AD, long-term continued operational safety is adequately assured by doing the repetitive inspections to detect discrepancies before they represent a hazard to the airplane, and by doing repairs within the specified time limits.

Differences Between Proposed AD, Canadian Airworthiness Directive, and Service Information

The service bulletin and Canadian airworthiness directive refer only to a "visual inspection" for discrepancies of the rear spar fittings. We have determined that the procedures in the service bulletin should be described as a "detailed inspection." Note 1 has been included in this proposed AD to define this type of inspection.

Although the service bulletin specifies to submit certain information to the manufacturer, this proposed AD does not include such a requirement.

The applicability specified in the service bulletin and Canadian airworthiness directive includes Model DHC-8-314 airplanes; however, those airplanes are not U.S. type certificated and are not included in the applicability in this proposed AD.

Cost Impact

The FAA estimates that 218 airplanes of U.S. registry would be affected by this proposed AD.

It would take about 16 work hours per rear spar fitting (two fittings per airplane) to accomplish the proposed inspection, at an average labor rate of \$65 per work hour. Based on these figures, the cost impact of the inspection proposed by this AD on U.S. operators is estimated to be \$453,440, or \$2,080 per airplane, per inspection cycle.

The cost impact figure discussed above is based on assumptions that no operator has yet done any of the proposed requirements of this AD action, and that no operator would do those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to do the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

The optional terminating modification, if done, would take about 16 work hours, at an average labor rate of \$65 per work hour. Required parts would cost about \$365 per airplane. Based on these figures, we estimate the cost of the optional terminating modification to be \$1,405 per airplane.

Regulatory Impact

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this proposal would not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption

ADDRESSES.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Bombardier, Inc. (Formerly de Havilland, Inc.): Docket 2003-NM-154-AD.

Applicability: Model DHC-8-102, -103, -106, -201, -202, -301, -311, and -315 series airplanes; certificated in any category; as listed in Bombardier Service Bulletin 8-27-83, Revision "A", dated February 8, 2002.

Compliance: Required as indicated, unless accomplished previously. To find and fix damage and prevent subsequent failure of the rear spar fittings between the flex shaft of the flap secondary drive and the wing-to-fuselage structure, which could result in loss of the wing, accomplish the following:

Repetitive Inspections/Corrective Action

(a) For airplanes with rear spar fittings having part number (P/N) 85320053,

85322060, or 85334180: Within 12 months after the effective date of this AD; do a detailed inspection for discrepancies (chafing, wear damage, cracking) of the rear spar fittings located between the flex shaft of the flap secondary drive and the wing-to-fuselage structure. Do the inspection as defined in Parts III.A., III.B., and III.D. of the Accomplishment Instructions of Bombardier Service Bulletin 8-27-83, Revision "A", dated February 8, 2002; except where the service bulletin specifies to report inspection findings, this AD does not require such reporting. Do the inspection per the service bulletin, and repeat the inspection thereafter at the applicable time specified in Part I.D. "Compliance" of the service bulletin. Any applicable corrective action (high frequency eddy current inspection for cracking, blending out wear damage, replacement of rear spar fittings) must be done at the applicable time specified in Part I.D. "Compliance" of the service bulletin.

Note 1: For the purposes of this AD, a detailed inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

Optional Terminating Modification

(b) Modification of the flex shaft of the flap secondary drive per Part III.C. of the Accomplishment Instructions of Bombardier Service Bulletin 8-27-83, Revision "A", dated February 8, 2002, terminates the repetitive inspections required by paragraph (a) of this AD.

Actions Done per Previous Issue of Service Bulletins

(c) Accomplishment of the inspections or the modification before the effective date of this AD in accordance with Bombardier Service Bulletin 8-27-83, dated October 19, 2001, is considered acceptable for compliance with the applicable actions specified in this AD.

Alternative Methods of Compliance

(d) In accordance with 14 CFR 39.19, the Manager, New York Aircraft Certification Office, FAA, is authorized to approve alternative methods of compliance for this AD.

Note 2: The subject of this AD is addressed in Canadian airworthiness directive CF-2001-42, dated November 23, 2001.

Issued in Renton, Washington, on November 21, 2003.

Kalene C. Yanamura,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-29698 Filed 11-26-03; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002-NM-292-AD]

RIN 2120-AA64

Airworthiness Directives; McDonnell Douglas Model MD-11 and MD-11F Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain McDonnell Douglas Model MD-11 and MD-11F airplanes. This proposal would require repetitive inspections of the transfer pipe assembly installation for the tail tank for damage and cracks, and corrective action, if necessary. This action is necessary to detect and correct damage and cracks to the transfer pipe assembly installation for the tail tank, which could result in fuel leakage and possible ignition. This action is intended to address the identified unsafe condition. **DATES:** Comments must be received by January 12, 2004.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2002-NM-292-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227-1232. Comments may also be sent via the Internet using the following address: 9-anm-nprmcomment@faa.gov. Comments sent via fax or the Internet must contain "Docket No. 2002-NM-292-AD" in the subject line and need not be submitted in triplicate. Comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 or 2000 or ASCII text.

The service information referenced in the proposed rule may be obtained from Boeing Commercial Aircraft Group, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1-L5A (D800-0024). This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at

the FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California.

FOR FURTHER INFORMATION CONTACT: Samuel S. Lee, Aerospace Engineer, Propulsion Branch, ANM-140L, FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California 90712-4137; telephone (562) 627-5338; fax (562) 627-5210.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this action may be changed in light of the comments received.

Submit comments using the following format:

- Organize comments issue-by-issue. For example, discuss a request to change the compliance time and a request to change the service bulletin reference as two separate issues.
- For each issue, state what specific change to the proposed AD is being requested.
- Include justification (e.g., reasons or data) for each request.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this action must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 2002-NM-292-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2002-NM-292-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.