

period and will consider any modifications that appear appropriate in adopting these rules as final with the concurrence and co-signature of the Office of Government Ethics.

Regulatory Flexibility Act

As Deputy General Counsel of the NEH, I have determined under the Regulatory Flexibility Act (5 U.S.C. chapter 6) that this regulation will not have a significant economic impact on a substantial number of small entities because it only affects NEH employees.

Paperwork Reduction Act

As Deputy General Counsel of the NEH, I have determined that the Paperwork Reduction Act (44 U.S.C. chapter 35) does not apply because these regulations do not contain any information collection requirements that require the approval of the Office of Management and Budget.

Unfunded Mandates Reform Act

For purposes of the Unfunded Mandates Reform Act of 1995 (2 U.S.C. chapter 25, subchapter II), this rule will not significantly or uniquely affect small governments and will not result in increased expenditures by State, local, and tribal governments, or by the private sector, of \$100 million or more (as adjusted for inflation) in any one year.

List of Subjects in 5 CFR Part 6601

Conflict of interests, Government employees, Standards of conduct.

Dated: August 13, 2003.

Michael McDonald,

Deputy General Counsel and Acting Designated Agency Ethics Officer, National Endowment for the Humanities.

Approved: August 27, 2003.

Amy L. Comstock,

Director, Office of Government Ethics.

■ For the reasons set forth in the preamble, the National Endowment for the Humanities, with the concurrence of the Office of Government Ethics, is amending title 5 of the Code of Federal Regulations by adding a new chapter LVI, consisting of part 6601, to read as follows:

CHAPTER LVI—NATIONAL ENDOWMENT FOR THE HUMANITIES

PART 6601—SUPPLEMENTAL STANDARDS OF ETHICAL CONDUCT FOR EMPLOYEES OF THE NATIONAL ENDOWMENT FOR THE HUMANITIES

Sec.

6601.101 General.

6601.102 Prior approval for outside employment.

Authority: 5 U.S.C. 7301, 5 U.S.C. App. (Ethics in Government Act of 1978); E.O. 12674, 54 FR 15159, 3 CFR, 1989 Comp., p. 215, as modified by E.O. 12731, 55 FR 42547, 3 CFR, 1990 Comp., p. 306; 5 CFR 2635.105, 2635.803.

§ 6601.101 General.

In accordance with 5 CFR part 2635.105, the regulations of this part apply to employees of the National Endowment for the Humanities (NEH) and supplement the Standards of Ethical Conduct for Employees of the Executive Branch contained in 5 CFR part 2635. In addition to the regulations in 5 CFR part 2635 and this part, employees of the NEH are subject to the executive branch employee responsibilities and conduct regulations at 5 CFR part 735, the executive branch financial disclosure regulations at 5 CFR part 2634, and the executive branch financial interests regulations at 5 CFR part 2640.

§ 6601.102 Prior approval for outside employment.

(a) Before engaging in any outside employment with a prohibited source within the meaning of 5 CFR 2635.203(d), whether or not for compensation, an employee other than a special Government employee must obtain written approval from his or her immediate supervisor and the Designated Agency Ethics Official. The request for approval shall include the following:

(1) A brief description of the employee's official duties, a brief description of the proposed outside employment (including the name of the person, group or other organization for whom the work is to be performed), and a brief description of the employee's discipline or inherent area of expertise based on experience or educational background; and

(2) Responses to the following questions:

(i) Whether the proposed outside employment will draw on non-public information or pertain to a matter to which the employee is presently assigned or has been assigned within the last year;

(ii) Whether the proposed outside employment pertains to an ongoing or announced agency policy or program;

(iii) Whether the proposed outside employment will involve teaching a course which is part of the established curriculum of an accredited institution of higher education, secondary school, elementary school, or an education or training program sponsored by a Federal, State or local government entity;

(iv) Whether the sponsor of the proposed outside employment has any interests before the NEH that may be substantially affected by the performance or nonperformance of the employee's duties;

(v) Whether the employee intends to refer to his or her official NEH position during the proposed outside employment, and, if so, the text of any disclaimers that he or she will use; and

(vi) Whether the employee will receive any payment or compensation for the proposed activity, and, if so, how much.

(b) Approval shall be granted only upon determination that the outside employment is not expected to involve conduct prohibited by statute or Federal regulation, including 5 CFR part 2635.

(c) Outside employment means any form of compensated or uncompensated non-Federal employment or business relationship involving the provision of personal services by the employee. It includes, but is not limited to, personal services such as acting as an officer, director, employee, agent, attorney, consultant, contractor, general partner, trustee, teacher or speaker. It includes writing done under arrangement with another person for production or publication of any written product.

[FR Doc. 03-22654 Filed 9-4-03; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 25

[Docket No. NM248; Special Conditions No. 25-241-SC]

Special Conditions: Embraer Model ERJ-170 series airplanes; Electronic Flight Control Systems; Automatic Takeoff Thrust Control System

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final special conditions.

SUMMARY: These special conditions are issued for the Embraer Model ERJ-170 series airplanes. These airplanes will have novel or unusual design features when compared to the state of technology envisioned in the airworthiness standards for transport category airplanes. These design features are associated with (1) Electronic Flight Control Systems and (2) Automatic Takeoff Thrust Control System (ATTCS). The applicable airworthiness regulations do not contain adequate or appropriate safety standards

for these design features. These special conditions contain the additional safety standards that the Administrator considers necessary to establish a level of safety equivalent to that established by the existing airworthiness standards. Additional special conditions will be issued for other novel or unusual design features of Embraer Model 170 series airplanes.

EFFECTIVE DATE: August 15, 2003.

FOR FURTHER INFORMATION CONTACT: Tom Groves, FAA, International Branch, ANM-116, Transport Airplane Directorate, Aircraft Certification Service, 1601 Lind Avenue SW., Renton, Washington 98055-4056; telephone (425) 227-1503; facsimile (425) 227-1149; email tom.groves@faa.gov.

SUPPLEMENTARY INFORMATION:

Background

On May 20, 1999, Embraer applied for a type certificate for its new Model ERJ-170 airplane. Two basic versions of the Model ERJ-170 are included in the application. The ERJ-170-100 airplane is a 69-78 passenger, twin-engine regional jet with a maximum takeoff weight of 81,240 pounds. The ERJ-170-200 is a derivative with a lengthened fuselage. Passenger capacity for the ERJ-170-200 is increased to 86, and maximum takeoff weight is increased to 85,960 pounds.

Type Certification Basis

Under the provisions of 14 CFR 21.17, Embraer must show that the Model ERJ-170 series airplanes meet the applicable provisions of 14 CFR part 25, as amended by Amendments 25-1 through 25-98.

If the Administrator finds that the applicable airworthiness regulations (*i.e.*, part 25, as amended) do not contain adequate or appropriate safety standards for Embraer Model ERJ-170 series airplanes because of novel or unusual design features, special conditions are prescribed under the provisions of § 21.16.

In addition to the applicable airworthiness regulations and special conditions, Embraer Model ERJ-170 series airplanes must comply with the fuel vent and exhaust emission requirements of 14 CFR part 34 and the noise certification requirements of 14 CFR part 36, and the FAA must issue a finding of regulatory adequacy pursuant to § 611 of Public Law 93-574, the "Noise Control Act of 1972."

Special conditions, as defined in 14 CFR 11.19, are issued in accordance with § 11.38 and become part of the type certification basis in accordance with

§ 21.17(a)(2), Amendment 21-69, effective September 16, 1991.

Special conditions are initially applicable to the model for which they are issued. Should the type certificate for that model be amended later to include any other model that incorporate the same novel or unusual design feature, or should any other model already included on the same type certificate be modified to incorporate the same novel or unusual design features, the special conditions would also apply to the other model under the provisions of § 21.101(a)(1), Amendment 21-69, effective September 16, 1991.

Novel or Unusual Design Features

The Embraer Model ERJ-170 series airplanes will incorporate the following novel or unusual design features:

I. Electronic Flight Control System

In airplanes with electronic flight control systems, there may not always be a direct correlation between pilot control position and the associated airplane control surface position. Under certain circumstances, a commanded maneuver that does not require a large control input may require a large control surface movement, possibly encroaching on a control surface or actuation system limit without the flightcrew's knowledge. This situation can arise in either manually piloted or autopilot flight and may be further exacerbated on airplanes where the pilot controls are not back-driven during autopilot system operation. Unless the flightcrew is made aware of excessive deflection or impending control surface limiting, control of the airplane by the pilot or autoflight system may be inadvertently continued so as to cause loss of control of the airplane or other unsafe characteristics of stability or performance.

Given these possibilities, a special condition for Embraer Model ERJ-170 series airplanes addresses control surface position awareness. This special condition requires that suitable display or annunciation of flight control position be provided to the flightcrew when near full surface authority (not crew-commanded) is being used, unless other existing indications are found adequate or sufficient to prompt any required crew actions. Suitability of such a display or annunciation must take into account that some piloted maneuvers may demand the airplane's maximum performance capability, possibly associated with a full control surface deflection. Therefore, simple display systems—that would function in both intended and unexpected control-

limiting situations—must be properly balanced to provide needed crew awareness and minimize nuisance alerts. A monitoring system that compares airplane motion, surface deflection, and pilot demand could be useful in eliminating nuisance alerting.

II. Automatic Takeoff Thrust Control System (ATTCS)

The Embraer Model ERJ-170 series airplane will incorporate an Automatic Takeoff Thrust Control System (ATTCS) in the engine's Full Authority Digital Electronic Control (FADEC) system architecture. The manufacturer requested that the FAA issue special conditions to allow performance credit to be taken for use of this function during go-around to show compliance with the requirement of § 25.121(d) regarding the approach climb gradient.

Section 25.904 and Appendix I refer to operation of ATTCS only during takeoff. Model ERJ-170 series airplanes have this feature for go-around also. The ATTCS will automatically increase thrust to the maximum go-around thrust available under the ambient conditions in the following circumstances:

- If an engine failure occurs during an all-engines-operating go-around, or
- If an engine has failed or been shut down earlier in the flight.

This maximum go-around thrust is the same as that used to show compliance with the approach-climb-gradient requirement of § 25.121(d). If the ATTCS is not operating, selection of go-around thrust will result in a lower thrust level.

The part 25 standards for ATTCS, contained in § 25.904 [Automatic takeoff thrust control system (ATTCS) and Appendix I], specifically restrict performance credit for ATTCS to takeoff. Expanding the scope of the standards to include other phases of flight, such as go-around, was considered when the standards were issued but was not accepted because of the effect on the flightcrew's workload. As stated in the preamble to amendment 25-62:

In regard to ATTCS credit for approach climb and go-around maneuvers, current regulations preclude a higher thrust for the approach climb [§ 25.121(d)] than for the landing climb [§ 25.119]. The workload required for the flightcrew to monitor and select from multiple in-flight thrust settings in the event of an engine failure during a critical point in the approach, landing, or go-around operations is excessive. Therefore, the FAA does not agree that the scope of the amendment should be changed to include the use of ATTCS for anything except the takeoff phase." (Refer to 52 FR 43153, November 9, 1987.)

The ATTCS incorporated on Embraer Model ERJ-170 series airplanes allows the pilot to use the same power setting procedure during a go-around, regardless of whether or not an engine fails. In either case, the pilot obtains go-around power by moving the throttles into the forward (takeoff/go-around) throttle detent. Since the ATTCS is permanently armed for the go-around phase, it will function automatically following an engine failure and advance the remaining engine to the ATTCS thrust level. This design adequately addresses the concerns about pilot workload which were discussed in the preamble to Amendment 25-62.

The system design allows the pilot to enable or disable the ATTCS function for takeoff. If the pilot enables ATTCS, a white "ATTCS" icon will be displayed on the Engine Indication and Crew Alerting System (EICAS) beneath the thrust mode indication on the display. This white icon indicates to the pilot that the ATTCS function is enabled. When the throttle lever is put in the TO/GA (takeoff/go-around) detent position, the white icon turns green, indicating to the pilot that the ATTCS is armed. If the pilot disables the ATTCS function for takeoff, no indication appears on the EICAS.

Regardless of whether the ATTCS is enabled for takeoff, it is automatically enabled when the airplane reaches the end of the take-off phase (that is, the thrust lever is below the TO/GA position and the altitude is greater than 1,700 feet above the ground, 5 minutes have elapsed since lift-off, or the airplane speed is greater than 140 knots).

During climb, cruise and descent, when the throttle is not in the TO/GA position, the ATTCS indication is inhibited. During descent and approach to land, until the thrust management system go-around mode is enabled—either by crew action or automatically when the landing gear are down and locked and flaps are extended—the ATTCS indication remains inhibited.

When the go-around thrust mode is enabled, unless the ATTCS system has failed, the white "ATTCS" icon will again be shown on the EICAS, indicating to the pilot that the system is enabled and in an operative condition in the event a go-around is necessary. If the thrust lever is subsequently placed in the TO/GA position, the ATTCS icon turns green, indicating that the system is armed and ready to operate.

If an engine fails during the go-around or during a one-engine-inoperative go-around in which an engine had been shut down or otherwise made inoperative earlier in the flight, the

EICAS indication will be GA RSV (go-around reserve) when the thrust levers are placed in the TO/GA position. The GA RSV indication means that the maximum go-around thrust under the ambient conditions has been commanded.

The propulsive thrust used to determine compliance with the approach climb requirements of § 25.121(d) is limited to the lesser of (i) the thrust provided by the ATTCS system, or (ii) 111 percent of the thrust resulting from the initial thrust setting with the ATTCS system failing to perform its uptrim function and without action by the crew to reset thrust. This requirement limits the adverse performance effects of a failure of the ATTCS and ensures adequate all-engines-operating go-around performance.

These special conditions require a showing of compliance with the provisions of § 25.904 and Appendix I applicable to the approach climb and go-around maneuvers.

The definition of a critical time interval for the approach climb case is of primary importance. During this time it must be extremely improbable to violate a flight path derived from the gradient requirement of § 25.121(d). That gradient requirement implies a minimum one-engine-inoperative flight path with the airplane in the approach configuration. The engine may have been inoperative before initiating the go-around, or it may become inoperative during the go-around. The definition of the critical time interval must consider both possibilities.

Discussion of Comments

Notice of proposed special conditions No. 25-03-03-SC for the Embraer Model ERJ-170 series airplane was published in the **Federal Register** on April 23, 2003 (68 FR 19958) and a Supplemental notice of proposed special conditions was published on June 5, 2003 (68 FR 33659). No comments were received after publication of the initial notice or the supplemental notice, and the special conditions are adopted as proposed.

Applicability

As discussed above, these special conditions are applicable to the Embraer Model ERJ-170 series airplanes. Should Embraer apply at a later date for a change to the type certificate to include another model incorporating the same novel or unusual design features, these special conditions would apply to that model as well under the provisions of § 21.101(a)(1), Amendment 21-69, effective September 16, 1991.

Under standard practice, the effective date of final special conditions would be 30 days after the date of publication in the **Federal Register**; however, as the certification date for the Embraer Model ERJ-170 series airplane is imminent, the FAA finds that good cause exists to make these special conditions effective upon issuance.

Conclusion

This action affects only certain novel or unusual design features on the Embraer Model ERJ-170 series airplanes. It is not a rule of general applicability, and it affects only the applicant who applied to the FAA for approval of these features on the airplane.

List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

■ The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701, 44702, 44704.

The Special Conditions

■ Accordingly, the Federal Aviation Administration (FAA) issues the following special conditions as part of the type certification basis for Embraer Model ERJ-170 series airplanes.

I. Electronic Flight Control System

In addition to compliance with §§ 25.143, 25.671 and 25.672, when a flight condition exists where, without being commanded by the crew, control surfaces are coming so close to their limits that return to the normal flight envelope and (or) continuation of safe flight requires a specific crew action, a suitable flight control position annunciation shall be provided to the crew, unless other existing indications are found adequate or sufficient to prompt that action.

Note: The term suitable also indicates an appropriate balance between nuisance and necessary operation.

II. Automatic Takeoff Thrust Control System (ATTCS)

To use the thrust provided by the ATTCS to determine the approach climb performance limitations, the Embraer Model ERJ-170 series airplane must comply with the requirements of § 25.904 and Appendix I, including the following requirements pertaining to the go-around phase of flight:

1. Definitions

(a) *TOGA—(Take Off/Go-Around)*. Throttle lever in takeoff or go-around position.

(b) *Automatic Takeoff Thrust Control System—(ATTCS)*. The Embraer Model ERJ-170 series ATTCS is defined as the entire automatic system available in takeoff when selected by the pilot and always in go-around mode; including all devices, both mechanical and electrical, that sense engine failure, transmit signals, and actuate fuel controls or power levers or increase engine power by other means on operating engines to achieve scheduled thrust or power increases and to furnish cockpit information on system operation.

(c) *Critical Time Interval*. The definition of the Critical Time Interval in appendix I, § I25.2(b) shall be expanded to include the following:

(1) When conducting an approach for landing using ATTCS, the critical time interval is defined as 120 seconds. A shorter time interval may be used if justified by a rational analysis. An accepted analysis that has been used on past aircraft certification programs is as follows:

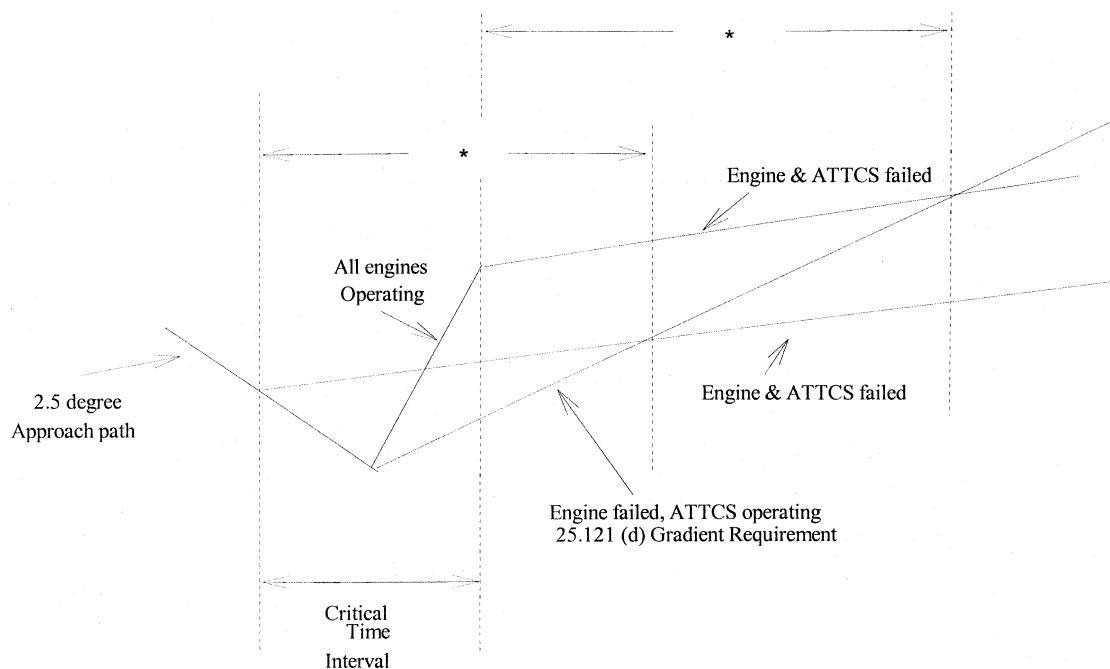
(i) The critical time interval begins at a point on a 2.5 degree approach glide path from which, assuming a simultaneous engine and ATTCS failure, the resulting approach climb flight path intersects a flight path originating at a later point on the same approach path corresponding to the part 25 one-engine-inoperative approach climb gradient. The period of time from the point of simultaneous engine and ATTCS failure to the intersection of these flight paths must be no shorter than the time interval used in evaluating the critical time interval for takeoff, beginning from the point of simultaneous engine and ATTCS failure and ending upon reaching a height of 400 feet.

(ii) The critical time interval ends at the point on a minimum performance, all-engines-operating go-around flight path from which, assuming a simultaneous engine and ATTCS failure, the resulting minimum approach climb flight path intersects a

flight path corresponding to the part 25 minimum one-engine-inoperative approach-climb-gradient. The all-engines-operating go-around flight path and the part 25 one-engine-inoperative, approach-climb-gradient flight path originate from a common point on a 2.5 degree approach path. The period of time from the point of simultaneous engine and ATTCS failure to the intersection of these flight paths must be no shorter than the time interval used in evaluating the critical time interval for the takeoff beginning from the point of simultaneous engine and ATTCS failure and ending upon reaching a height of 400 feet.

(2) The critical time interval must be determined at the altitude resulting in the longest critical time interval for which one-engine-inoperative approach climb performance data are presented in the Airplane Flight Manual (AFM).

(3) The critical time interval is illustrated in the following figure:



The engine and ATTCS failed time interval must be no shorter than the time interval from the point of simultaneous engine and ATTCS failure to a height of 400 feet used to comply with I25.2(b) for ATTCS use during takeoff.

2. Performance and System Reliability Requirements.

The applicant must comply with the following performance and ATTCS reliability requirements:

(a) An ATTCS failure or combination of failures in the ATTCS during the critical time interval:

(1) Shall not prevent the insertion of the maximum approved go-around thrust or power or must be shown to be an improbable event.

(2) Shall not result in a significant loss or reduction in thrust or power or must be shown to be an extremely improbable event.

(b) The concurrent existence of an ATTCS failure and an engine failure

during the critical time interval must be shown to be extremely improbable.

(c) All applicable performance requirements of part 25 must be met with an engine failure occurring at the most critical point during go-around with the ATTCS system functioning.

(d) The probability analysis must include consideration of ATTCS failure occurring after the time at which the flightcrew last verifies that the ATTCS is in a condition to operate until the beginning of the critical time interval.

(e) The propulsive thrust obtained from the operating engine after failure of the critical engine during a go-around used to show compliance with the one-engine-inoperative climb requirements of § 25.121(d) may not be greater than the lesser of:

(i) The actual propulsive thrust resulting from the initial setting of power or thrust controls with the ATTCS functioning; or

(ii) 111 percent of the propulsive thrust resulting from the initial setting of power or thrust controls with the ATTCS failing to reset thrust or power and without any action by the crew to reset thrust or power.

3. Thrust Setting.

(a) The initial go-around thrust setting on each engine at the beginning of the go-around phase may not be less than any of the following:

(1) That required to permit normal operation of all safety-related systems and equipment dependent upon engine thrust or power lever position; or

(2) That shown to be free of hazardous engine response characteristics when thrust or power is advanced from the initial go-around position to the maximum approved power setting.

(b) For approval of an ATTCS for go-around, the thrust setting procedure must be the same for go-arounds initiated with all engines operating as for go-arounds initiated with one engine inoperative.

4. Powerplant Controls.

(a) In addition to the requirements of § 25.1141, no single failure or malfunction, or probable combination thereof, of the ATTCS, including associated systems, may cause the failure of any powerplant function necessary for safety.

(b) The ATTCS must be designed to accomplish the following:

(1) Following any single engine failure during go around: Apply thrust or power on the operating engine(s) to achieve the maximum approved go-around thrust without exceeding engine operating limits;

(2) Permit manual decrease or increase in thrust or power up to the maximum go-around thrust approved for the airplane under existing conditions through the use of the power lever. For airplanes equipped with limiters that automatically prevent engine operating limits from being exceeded under existing ambient conditions, other means may be used to increase the thrust in the event of an ATTCS failure. Any such means must be located on or forward of the power levers; be easily identified and operated under all operating conditions by a single action of either pilot with the

hand that is normally used to actuate the power levers, and meet the requirements of § 25.777 (a), (b), and (c);

(3) Provide a means to verify to the flightcrew before beginning an approach for landing that the ATTCS is in a condition to operate (unless it can be demonstrated that an ATTCS failure combined with an engine failure during an entire flight is extremely improbable); and

(4) Provide a means for the flightcrew to deactivate the automatic function. This means must be designed to prevent inadvertent deactivation.

5. In addition to the requirements of § 25.1305, the following requirements pertaining to powerplant instruments must be met:

(a) A means must be provided to indicate when the ATTCS is in the armed or ready condition; and

(b) If the inherent flight characteristics of the airplane do not provide adequate warning that an engine has failed, a warning system that is independent of the ATTCS must be provided to give the pilot a clear warning of any engine failure during go-around.

Issued in Renton, Washington, on August 15, 2003.

Kyle Olsen,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

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BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001-NM-187-AD; Amendment 39-13293; AD 2003-18-02]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A330 and A340 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Airbus Model A330 and A340 series airplanes, that requires, among other actions, a detailed inspection of the rudder travel limitation unit for proper adjustment, measurement of the desynchronization of rudder servo-controls, installation of rigging placards for rudder servo-controls, and follow-on and corrective actions if necessary. This action is necessary to prevent desynchronization of the rudder servo-controls, which

could result in high load factors on the rudder servo-controls, and consequent reduced structural integrity of the attachment fittings for the rudder servo-controls. This action is intended to address the identified unsafe condition.

DATES: Effective October 10, 2003.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of October 10, 2003.

ADDRESSES: The service information referenced in this AD may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2125; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Airbus Model A330 and A340 series airplanes was published in the **Federal Register** on May 29, 2003 (68 FR 31991). That action proposed to require, among other actions, a detailed inspection of the rudder travel limitation unit for proper adjustment, measurement of the desynchronization of rudder servo-controls, installation of rigging placards for rudder servo-controls, and follow-on and corrective actions if necessary.

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comments received.

Request To Extend Compliance Time

Two commenters request that the compliance time of the proposed AD be extended from 16 months to 18 months. One of the commenters is the operator of the nine U.S.-registered Airbus Model A330 airplanes affected by the proposed AD. Extension of the compliance time to 18 months would match the Airbus C-check interval. The second commenter supports the comments of the first commenter and offers no additional comment.

The FAA does not agree with the commenters' request to extend the