

Actions	Compliance	Procedures
(1) Check the airplane logbook: (i) For all affected airplanes: to determine if Modifications H 225, H 269, and H 360 are incorporated; and (ii) For only these airplanes that incorporate Modification H 197 (glider towing capabilities): to determine if Modification H 275 is incorporated.	Within the next 90 days after April 25, 2003 (the effective date of this AD).	The owner/operator holding at least a private pilot certificate as authorized by section 43.7 of the Federal Aviation Regulations (14 CFR 43.7) may check the airplane logbook.
(2) If, by checking the airplane logbook, you can positively determine that all the applicable modifications in paragraphs (d)(1)(i) and (d)(1)(ii) are incorporated, you must make an entry into the aircraft records that shows compliance with paragraphs (d)(1) and (d)(2) of this AD in accordance with section 43.9 of the Federal Aviation Regulations (14 CFR 43.9).	Not applicable.	the owner/operator holding at least a private pilot certificate is authorized by section 43.7 of the Federal Aviation Regulations (14 CFR 43.7) may check the airplane logbook.
(3) If, by checking the airplane logbook, you determine that all the applicable modifications in paragraphs (d)(1)(i) and (d)(1)(ii) are not incorporated, or you cannot positively show that they are incorporated: (i) Incorporate each missing modification; and (ii) you must make an entry into the aircraft records that shows compliance with this portion of the AD in accordance with section 43.9 of the Federal Aviation Regulations (14 CFR 43.9).	Within the next 90 days after April 25, 2003 (the effective date of this AD), unless already accomplished.	British Aerospace Aerostructures Limited has issued BAe Aircraft Technical News Sheet CT (C1) No. 200, Issue 1, dated March 1, 1997.
(4) Do not incorporate Modification H 197 unless Modification H 275 has also been incorporated.	As of April 25, 2003 the (effective date of this AD).	British Aerospace Aerostructures Limited has issued BAe Aircraft Technical News Sheet CT (C1) No. 200, Issue 1, dated March 1, 1997.

Note 2: Although not required by this AD, FAA highly recommends you incorporate Modification H 282.

(e) *Can I comply with this AD in any other way?* You may use an alternative method of compliance or adjust the compliance time if:

- (1) Your alternative method of compliance provides an equivalent level of safety; and
- (2) The Manager, Atlanta Aircraft Certification Office (ACO), approves your alternative. Submit your request through an FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Atlanta ACO.

Note 3: This AD applies to each airplane identified in paragraph (a) of this AD, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if you have not eliminated the unsafe condition, specific actions you propose to address it.

(f) *Where can I get information about any already-approved alternative methods of compliance?* Contact Cindy Lorenzen, Aerospace Engineer, FAA, Atlanta Aircraft Certification Office, 1895 Phoenix Boulevard, Suite 450, Atlanta, Georgia; telephone: (770) 703-6078; facsimile: (770) 703-6097.

(g) *What if I need to fly the airplane to another location to comply with this AD?* The FAA can issue a special flight permit under sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and

21.199) to operate your airplane to a location where you can accomplish the requirements of this AD.

(h) *Are any service bulletins incorporated into this AD by reference?* Actions required by this AD must be done in accordance with BAe Aircraft Technical News Sheet CT (C1) No 200, Issue 1, dated March 1, 1997. The Director of the Federal Register approved this incorporation by reference under 5 U.S.C. 552(a) and 1 CFR part 51. You may get copies from DeHavilland Support Limited, Duxford Airfield, Bldg. 213, Cambridgeshire, CB2 4QR, United Kingdom, telephone: +44 1223 830090, facsimile: +44 1223 830085, e-mail: info@dhsupport.com. You may view copies at the FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri, or at the Office of the Federal Register, 800 North Capitol Street, NW, suite 700, Washington, DC.

(i) *When does this amendment become effective?* This amendment becomes effective on April 25, 2003.

Issued in Kansas City, Missouri, on March 4, 2003.

Dorenda D. Baker,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-6040 Filed 3-17-03; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2000-CE-63-AD; Amendment 39-13081; AD 2003-05-05]

RIN 2120-AA64

Airworthiness Directives; Robert E. Rust Models DeHavilland DH.C1 Chipmunk 21, 22, and 22A Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD) that applies to certain Robert E. Rust (R.E. Rust) Models DeHavilland DH.C1 Chipmunk 21, 22, and 22A airplanes. This AD requires you to inspect the fuselage to determine if a steel fuselage center-section tie bar fitted with bushings in the end lug bolt holes is installed. If this bushed steel fuselage center-section tie bar is installed, this AD decreases the safe life limit. This AD is the result of reports that certain replacement steel fuselage center-section tie bars installed on the affected airplanes could fail before the originally published safe life limit. The actions specified by this AD are intended to prevent early failure of these bushed steel fuselage center-section tie bars, which could result in reduced structural

integrity of the wings. Such a condition could lead to loss of control of the airplane.

DATES: This AD becomes effective on April 25, 2003.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in the regulations as of April 25, 2003.

ADDRESSES: You may get the service information referenced in this AD from DeHavilland Support Limited, Duxford Airfield, Bldg. 213, Cambridgeshire, CB2 4QR, United Kingdom, telephone: +44 1223 830090, facsimile: +44 1223 830085, e-mail: *info@dhsupport.com*. You may view this information at the Federal Aviation Administration (FAA), Central Region, Office of the Regional Counsel, Attention: Rules Docket No. 2000-CE-63-AD, 901 Locust, Room 506, Kansas City, Missouri 64106; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Cindy Lorenzen, Aerospace Engineer, FAA, Atlanta Aircraft Certification Office, 1895 Phoenix Boulevard, Suite 450, Atlanta, Georgia; telephone: (770) 703-6078; facsimile: (770) 703-6097.

SUPPLEMENTARY INFORMATION:

Discussion

What Events Have Caused This AD?

The FAA has received reports that an unsafe condition may exist on certain R.E. Rust Models DeHavilland DH.C1 Chipmunk 21, 22, and 22A airplanes. After a review of several of these airplanes, we have determined that steel

fuselage center-section tie bars, part number RD.C1.FS.107, are being installed as replacements parts. Some of these part numbers have been fitted with bushings in the end lugs to cover scored or oversized holes.

The use of bushings in the end of the lugs on these parts severely reduces the safe life limit. The original safe life limit established for the steel fuselage center-section tie bar was 30,000 fatigue hours. Fatigue hours are hours time-in-service multiplied by the role factor (operational use).

What Is the Potential Impact if FAA Took No Action?

This condition, if not corrected, could result in failure of the steel fuselage center-section tie bar. Such failure could lead to loss of control of the airplane.

Has FAA Taken Any Action to This Point?

We issued a proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an AD that would apply to certain R.E. Rust Models DeHavilland DH.C1 Chipmunk 21, 22, and 22A airplanes. This proposal was published in the **Federal Register** as a notice of proposed rulemaking (NPRM) on November 15, 2002 (67 FR 69149). The NPRM proposed to require you to check the airplane logbook to determine if a steel fuselage center-section tie bar, part number RD.C1.FS.107, is installed on the airplane. If this part number is installed, the NPRM proposed to require you to inspect the end lugs to determine if bushings are installed in the bolt holes. If bushings are present, the NPRM

also proposed to reduce the safe life of that part from 30,000 fatigue hours to 16,000 fatigue hours.

Was the Public Invited To Comment?

The FAA encouraged interested persons to participate in the making of this amendment. We did not receive any comments on the proposed rule or on our determination of the cost to the public.

FAA's Determination

What Is FAA's Final Determination on This Issue?

After careful review of all available information related to the subject presented above, we have determined that air safety and the public interest require the adoption of the rule as proposed except for minor editorial corrections. We have determined that these minor corrections:

- Provide the intent that was proposed in the NPRM for correcting the unsafe condition; and
- do not add any additional burden upon the public than was already proposed in the NPRM.

Cost Impact

How Many Airplanes Does This AD Impact?

We estimate that this AD affects 54 airplanes in the U.S. registry.

What Is the Cost Impact of This AD on Owners/Operators of the Affected Airplanes?

We estimate the following costs to accomplish the inspection:

Labor cost	Parts cost	Total cost per airplane	Total cost on U.S. operators
12 workhours × \$60 per hour = \$720	No parts required	\$720	\$720 × 54 = \$38,880.

We estimate the following costs to accomplish any necessary replacements that will be required based on the

results of the inspection. We have no way of determining the number of

airplanes that may need such replacement:

Labor cost	Parts cost	Total cost per airplane
80 workhours × \$60 per hour = \$4,800	\$2,250	\$4,800 + \$2,250 = \$7,050.

Regulatory Impact

Does This AD Impact Various Entities?

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is

determined that this final rule does not have federalism implications under Executive Order 13132.

Does This AD Involve a Significant Rule or Regulatory Action?

For the reasons discussed above, I certify that this action (1) Is not a “significant regulatory action” under Executive Order 12866; (2) is not a

“significant rule” under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the final evaluation prepared for this action is contained in the Rules Docket. A copy

of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. FAA amends § 39.13 by adding a new AD to read as follows:

2003-05-05 Robert E. Rust: Amendment 39-13081; Docket No. 2000-CE-63-AD.

(a) *What airplanes are affected by this AD?* This AD affects R.E. Rust Models DeHavilland DH.C1 Chipmunk 21, 22, and 22A airplanes, serial numbers C1-001 through C1-1014, that are type certificated in any category.

Note 1: We recommend all owners/operators of DeHavilland DH.C1 Chipmunk 21, 22, and 22A airplanes, serial numbers C1-001 through C1-1014, with experimental airworthiness certificates comply with the actions required in this AD.

(b) *Who must comply with this AD?* Anyone who wishes to operate any of the

airplanes identified in paragraph (a) of this AD must comply with this AD.

(c) *What problem does this AD address?* The actions specified by this AD are intended to prevent failure of the steel fuselage center-section tie bar prior to the originally published safe life, which could result in reduced structural integrity of the wings. Such a condition could lead to loss of control of the airplane. Steel fuselage center-section tie bars fitted with bushings in the end lug bolt holes have a reduced safe life of 16,000 fatigue hours.

(d) *What actions must I accomplish to address this problem?* To address this problem, you must accomplish the following:

Actions	Compliance	Procedures
(1) Check the airplane logbook to determine if a steel fuselage center-section tie bar, part number (P/N) RD.C1.FS.107, is installed. Initial steel tie bar fitments were done under cover of Repair Drawings R.C1.FS.191 and RD.C1.FS.106. Later these drawings were included in Modification H.288 so fitment may be logged under either.	Upon accumulating 16,000 fatigue hours or within the next 100 hours time-in-service (TIS) after April 25, 2003 (the effective date of this AD), whichever occurs later.	The owner/operator holding at least a private pilot certificate as authorized by section 43.7 of the Federal Aviation Regulations (14 CFR 43.7) may check the airplane logbook. Calculate fatigue hours by multiplying the TIS by the role factor in accordance with British Aerospace Mandatory Technical News Sheet Series: Chipmunk (C1), No. 138, Issue: 5, dated August 1, 1985.
(2) If, by checking the airplane logbook, you can positively determine that a steel fuselage center-section tie bar, P/N RD.C1.FS.107, is not installed. (i) you must make an entry into the aircraft records that shows compliance with paragraphs (d)(1) and (d)(2) of this AD in accordance with section 43.9 of the Federal Aviation Regulations (14 CFR 43.9); and (ii) continue to comply with the published life limits of the installed tie bar.	Not applicable	The owner/operator holding at least a private pilot certificate as authorized by section 43.7 of the Federal Aviation Regulations (14 CFR 43.7) may check the airplane logbook.
(3) If, by checking the airplane logbook, you determine that a steel fuselage center-section tie bar, P/N RD.C1.FS.107, is installed, or cannot positively show that one is not installed. (i) inspect the lug bolt holes to determine if bushings have been installed. (ii) if bushings have been installed, the safe life limit for that part is now 16,000 fatigue hours;. (iii) if bushing have not been installed, the safe life limit for that part remains at 30,000 fatigue hours; and (iv) make an entry into the aircraft records that shows compliance with this portion of the AD in accordance with section 43.9 of the Federal Aviation Regulations (14 CFR 43.9).	Prior to further flight after the logbook check required in paragraph (d)(1) of this AD.	In accordance with British Aerospace Mandatory Technical News Sheet Series: Chipmunk (C1), No. 175, Issue 1, dated August 1, 1985.
(4) The following are the safe life limit for steel fuselage center-section tie bars, P/N RD.C1.FS.107. (i) If fitted with bushings in the end lug bolt holes: 16,000 fatigue hours; and (ii) If not fitted with bushings in the end lug bolt holes: 30,000 fatigue hours.	As of April 25, 2003 (the effective date of this AD).	Not applicable.

(e) *Can I comply with this AD in any other way?* You may use an alternative method of compliance or adjust the compliance time if:

- (1) Your alternative method of compliance provides an equivalent level of safety; and
- (2) The Manager, Atlanta Aircraft Certification Office (ACO), approves your

alternative. Submit your request through an FAA Principal Maintenance

Inspector, who may add comments and then send it to the Manager, Atlanta ACO.

Note: This AD applies to each airplane identified in paragraph (a) of this AD, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if you have not eliminated the unsafe condition, specific actions you propose to address it.

(f) *Where can I get information about any already-approved alternative methods of compliance?* Contact Cindy Lorenzen, Aerospace Engineer, FAA, Atlanta Aircraft Certification Office, 1895 Phoenix Boulevard, Suite 450, Atlanta, Georgia; telephone: (770) 703-6078; facsimile: (770) 703-6097.

(g) *What if I need to fly the airplane to another location to comply with this AD?* The FAA can issue a special flight permit under sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate your airplane to a location where you can accomplish the requirements of this AD.

(h) *Are any service bulletins incorporated into this AD by reference?* Actions required by this AD must be done in accordance with British Aerospace Mandatory Technical News Sheet Series: Chipmunk (C1), No. 138, Issue: 5, dated August 1, 1985, and British Aerospace Mandatory Technical News Sheet Series: Chipmunk (C1), No. 175, Issue 1, dated August 1, 1985. The Director of the Federal Register approved this incorporation by reference under 5 U.S.C. 552(a) and 1 CFR part 51. You may get copies from DeHavilland Support Limited, Duxford Airfield, Bldg. 213, Cambridgeshire, CB2 4QR, United Kingdom, telephone: +44 1223 830090, facsimile: +44 1223 830085, e-mail: info@dhsupport.com. You may view copies at the FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri, or at the Office of the Federal Register, 800 North Capitol Street, NW., Suite 700, Washington, DC.

(i) *When does this amendment become effective?* This amendment becomes effective on April 25, 2003.

Issued in Kansas City, Missouri, on March 4, 2003.

Dorenda D. Baker,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-6045 Filed 3-17-03; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002-NM-216-AD; Amendment 39-12912; AD 2002-21-06]

RIN 2120-AA64

Airworthiness Directives; McDonnell Douglas Model DC-9-81 (MD-81), DC-9-82 (MD-82), DC-9-83 (MD-83), DC-9-87 (MD-87), and MD-88 Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; correction.

SUMMARY: This document corrects information in an existing airworthiness directive (AD) that applies to all McDonnell Douglas Model DC-9-81 (MD-81), DC-9-82 (MD-82), DC-9-83 (MD-83), DC-9-87 (MD-87), and MD-88 airplanes. That AD currently requires revisions to the Airplane Flight Manual; installation of inspection aids on the wing upper surfaces; and, among other actions, installation of an overwing heater blanket system or primary upper wing ice detection system, and installation of a heater protection panel or an equipment protection device on certain overwing heater blanket systems. That AD also requires disabling the anti-ice systems for the upper wing surface on certain airplanes. This document corrects a reference to an incorrect paragraph. This correction is necessary to provide the correct paragraph reference.

DATES: Effective November 8, 2002.

The incorporation by reference of certain publications listed in the regulations was approved previously by the Director of the Federal Register as of November 8, 2002 (67 FR 65298, October 24, 2002).

The incorporation by reference of certain publications, as listed in the regulations, was approved previously by the Director of the Federal Register as January 17, 1992 (57 FR 2014, January 17, 1992).

The incorporation by reference of certain other publications, as listed in the regulations, was approved previously by the Director of the Federal Register as of May 7, 2001 (66 FR 17499, April 2, 2001).

FOR FURTHER INFORMATION CONTACT:

Technical Information: Cheyenne Del Carmen, Aerospace Engineer, Systems and Equipment Branch, ANM-130L, FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard,

Lakewood, California 90712-4137; telephone (562) 627-5338; fax (562) 627-5210.

Other Information: Judy Golder, Airworthiness Directive Technical Editor/Writer; telephone (425) 687-4241, fax (425) 227-1232. Questions or comments may also be sent via the Internet using the following address: judy.golder@faa.gov. Questions or comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 for Windows or ASCII text.

SUPPLEMENTARY INFORMATION: On October 9, 2002, the Federal Aviation Administration (FAA) issued AD 2002-21-06, amendment 39-12912 (67 FR 65298, October 24, 2002), which applies to all McDonnell Douglas Model DC-9-81 (MD-81), DC-9-82 (MD-82), DC-9-83 (MD-83), DC-9-87 (MD-87), and MD-88 airplanes. That AD requires revisions to the Airplane Flight Manual; installation of inspection aids on the wing upper surfaces; and, among other actions, installation of an overwing heater blanket system or primary upper wing ice detection system, and installation of a heater protection panel or an equipment protection device on certain overwing heater blanket systems. That AD also requires disabling the anti-ice systems for the upper wing surface on certain airplanes. The actions required by that AD are intended to prevent ice ingestion into one or both engines and consequent loss of thrust from one or both engines; and damage to the upper wing skin surface and its structure, due to prolonged short-circuit electrical arcing of certain anti-ice systems.

On December 23, 2002, the FAA issued AD 2002-21-06 COR, amendment 39-12912 (68 FR 5, January 2, 2003), that corrected certain references that were transposed in two paragraphs.

Need for the Correction

After publication of that correction, the FAA received comments from two operators that point out the need for further correction. One commenter notes that, in the corrected AD, the statement at the beginning of the body advising that AD 2002-21-06 supersedes AD 2001-06-16 COR, amendment 39-12163, is missing. The commenter states that adding that statement would clarify that AD 2001-06-16 has been superseded. Both