

What Airplanes Are Affected by This AD?

(c) This AD affects Model PC-12 and PC-12/45 airplanes, all serial numbers, that are:
 (1) equipped with an inboard and/or outboard flap flexshaft, part number (P/N) 945.02.02.203 and/or P/N 945.02.02.204; and
 (2) certificated in any category.

What Is the Unsafe Condition Presented in This AD?

(d) This AD is the result of mandatory continuing airworthiness information (MCAI) issued by the airworthiness authority for Switzerland. The actions specified in this AD are intended to prevent rupture of the flap flexshafts due to corrosion, which could cause the flap system to become inoperable.

What Must I Do To Address This Problem?

(e) To address this problem, you must do the following. If you already replaced both the inboard and outboard flap flexshafts, P/N 945.02.02.203 and P/N 945.02.02.204, following Pilatus PC12 Service Bulletin No. 27-015, dated June 4, 2003, then paragraph (e)(5) of this AD is the only paragraph that applies to you:

Actions	Compliance	Procedures
(1) <i>For affected airplanes with a manufacturer serial number (MSN) of 489 or lower:</i> check the airplane logbook to determine if P/N 945.02.02.203 and P/N 945.02.02.204 inboard and outboard flap flexshafts are installed.	Within the next 30 days after July 26, 2004 (the effective date of this AD).	The owner/operator holding at least a private pilot certificate as authorized by section 43.7 of the Federal Aviation Regulations (14 CFR 43.7) may perform this check.
(2) <i>For affected airplanes with a MSN of 490 and above:</i> check the airplane logbook to ensure that P/N 945.02.02.203 and P/N 945.02.02.204 inboard and outboard flap flexshafts have not been installed since delivery.	Within the next 30 days after July 26, 2004 (the effective date of this AD).	The owner/operator holding at least a private pilot certificate as authorized by section 43.7 of the Federal Aviation Regulations (14 CFR 43.7) may perform this check.
(3) If you can positively determine that both P/Ns 945.02.02.203 and 945.02.02.204 inboard and outboard flap flexshafts are not installed, then no replacement is required.	Not Applicable	Not applicable.
(4) If you cannot positively determine that both P/Ns 945.02.02.203 and 945.02.02.204, inboard and outboard flap flexshafts are not installed, then you must replace each one or both with P/N 945.02.02.205 and P/N 945.02.02.206, as applicable (or a later FAA-approved manufactured part of improved design).	Before further flight after the logbook checks required in paragraph (e)(1) and (e)(2) of this AD.	Follow Pilatus PC12 Service Bulletin No. 27-015 as specified in paragraph (f) of this AD.
(5) Do not install inboard and outboard flap flexshafts, P/Ns 945.02.02.203 and 945.02.02.204.	As of July 26, 2004 (the effective date of this AD).	Not applicable.

What Revision Levels Do the Affected Service Bulletin Incorporate?

(f) The service bulletin required to do the actions required in this AD incorporates the following pages:

Affected pages	Revision level	Date
1 and 2	A	November 13, 2003.
3 through 11	Original Issue	June 4, 2003.

May I Request an Alternative Method of Compliance?

(g) You may request a different method of compliance or a different compliance time for this AD by following the procedures in 14 CFR 39.19. Unless FAA authorizes otherwise, send your request to your principal inspector. The principal inspector may add comments and will send your request to the Manager, Standards Office, Small Airplane Directorate, FAA. For information on any already approved alternative methods of compliance, contact Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4059; facsimile: (816) 329-4090.

Does This AD Incorporate Any Material by Reference?

(h) You must do the actions required by this AD following the instructions in Pilatus PC12 Service Bulletin No. 27-015, pages 1 and 2, Revision A, dated November 13, 2003, pages 3 through 11, Original issue, dated June 4, 2003. The Director of the Federal Register approved the incorporation by reference of this service bulletin in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. You may get a copy from Pilatus Aircraft Ltd., Customer Liaison Manager, CH-6371 Stans, Switzerland; telephone: +41 41 619 6208; facsimile: +41 41 619 7311; e-mail: SupportPC12@pilatus-aircraft.com or from Pilatus Business Aircraft Ltd., Product Support Department, 11755 Airport Way, Broomfield, Colorado 80021; telephone: (303) 465-9099; facsimile: (303) 465-6040. You may review copies at FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri 64106; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Is There Other Information That Relates to This Subject?

(i) Swiss AD Number HB-2004-068, dated March 4, 2004, also addresses the subject of this AD.

Issued in Kansas City, Missouri, on June 3, 2004.

Dorenda D. Baker,
 Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 04-13334 Filed 6-15-04; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2000-NM-376-AD; Amendment 39-13666; AD 2004-12-07]

RIN 2120-AA64

Airworthiness Directives; Boeing Model 757 Series Airplanes Equipped With Rolls Royce RB211 Engines

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment supersedes an existing airworthiness directive (AD), applicable to certain Boeing Model 757 series airplanes equipped with Rolls Royce RB211 engines, that currently requires modification of the nacelle strut and wing structure. This amendment requires, for certain airplanes, repetitive detailed inspections of certain aft bulkhead fasteners for loose or missing fasteners, and corrective action if necessary. For certain other airplanes, this amendment requires a one-time detailed inspection of the middle gusset of the inboard side load fitting for proper alignment and realignment if necessary; a one-time eddy current inspection of certain fastener holes for cracking, and repair if necessary; and a detailed inspection of certain fasteners for loose or missing fasteners, and replacement with new fasteners if necessary. The actions specified by this AD are intended to prevent fatigue cracking in primary strut structure and consequent reduced structural integrity of the strut. These actions are intended to address the identified unsafe condition.

DATES: Effective July 21, 2004.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of July 21, 2004.

ADDRESSES: The service information referenced in this AD may be obtained from Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124-2207. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

FOR FURTHER INFORMATION CONTACT: Dennis Stremick, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 917-6450; fax (425) 917-6590.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) by superseding AD 99-24-07, amendment 39-11431 (64 FR 66370, November 26, 1999), which is applicable to certain Boeing Model 757 series airplanes equipped with Rolls

Royce RB211 engines, was published as a supplemental Notice of Proposed Rulemaking in the **Federal Register** on April 15, 2003 (68 FR 18170). The action proposed to continue to require modification of the nacelle strut and wing structure. The action proposed to require, for certain airplanes, repetitive detailed inspections of certain aft bulkhead fasteners for loose or missing fasteners, and corrective action if necessary. For certain other airplanes, the action proposed to require a one-time detailed inspection of the middle gusset of the inboard side load fitting for proper alignment and realignment if necessary; a one-time eddy current inspection of certain fasteners holes for cracking; and repair if necessary; and a detailed inspection of certain fasteners for loose or missing fasteners, and replacement with new fasteners if necessary. Additionally, the action proposed to require that certain actions specified in Boeing Service Bulletin 757-54-0035, Revision 2, dated June 13, 2002 (specified in the supplemental NPRM as one of the appropriate sources of service information), be done using Boeing-supplied tools.

Request for Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comments received.

Comments Received That Resulted in Changes to the AD

Requests To Extend Repetitive Inspection Interval

Several commenters request that the repetitive inspection interval of every six months proposed by paragraph (b) of the supplemental NPRM be extended. Two commenters state that repetitive inspection intervals of every 3,000 flight cycles would be less of a burden for the operators. Another commenter advises that the airplane manufacturer's intent was to require repetitive inspections every 12,000 flight cycles or 72 months, whichever occurs first.

The FAA concurs that the repetitive inspection interval may be extended somewhat. We understand that the manufacturer has recommended an interval of every 12,000 flight cycles or 72 months. However, we have recently received reports of field experience that show the fasteners can loosen in less than 72 months. Therefore, we have revised paragraph (b) of the AD to specify a repetitive inspection interval not to exceed 6,000 flight cycles or 36 months, whichever occurs first.

Requests To Clarify Paragraph (f) of the Supplemental NPRM

One commenter states that the way the supplemental NPRM is written, it would require the actions specified in paragraph (f) of the supplemental NPRM to be accomplished after each repetitive inspection required by paragraph (b) of the supplemental NPRM. Two commenters request that the supplemental NPRM be revised to clearly specify that accomplishment of the requirements of paragraph (f) of the supplemental NPRM is terminating action.

We agree with the need to clarify paragraph (f) of the supplemental NPRM. The requirements of paragraph (f) of the AD (to increase the diameter of the fastener holes and to install new fasteners) apply to those airplanes on which the actions specified in paragraph (d) of the AD have been accomplished. We have revised paragraph (f) of the AD to reflect that clarification. Additionally, we agree that the actions specified in paragraph (f) of this AD terminates the repetitive inspection requirements of this AD. We have revised the AD accordingly.

Comments Received That Resulted in No Change to the Supplemental NPRM

Requests To Withdraw Rulemaking Until New Service Information Is Issued

Several commenters request that the supplemental NPRM be withdrawn and that, instead, new rulemaking be proposed to specify that the initial inspection specified in paragraph (b) of the supplemental NPRM be accomplished within 90 days after the release of a new Boeing service bulletin (Boeing Alert Service Bulletin 757-54A0047). The commenters state that using the new service information would simplify and clarify the actions proposed in the supplemental NPRM.

We do not agree that this AD should be withdrawn. We have not reviewed or approved new service information specified by the commenters. In this case, we find that to withdraw this AD and initiate new proposed rulemaking (providing for public opportunity to comment) would significantly delay the rulemaking process and would be inappropriate in light of the identified unsafe condition. Therefore, no change is necessary to the AD in this regard. In the future, if the manufacturer elects to provide new service information, the service information can be evaluated and approved in accordance with paragraph (h) of this AD.

Request To Extend Compliance Time of Paragraph (b) of the Supplemental NPRM

One commenter requests that, for airplanes that have completed the modification specified in Boeing Service Bulletin 757-54-0035, the compliance time specified in paragraph (b) of the supplemental NPRM be extended. The commenter states that the compliance time should be extended because the previous modification was done on those airplanes in a shop environment.

We do not agree that extending the compliance time specified in paragraph (b) of this AD is necessary. The requirements of paragraph (b) of this AD apply only to airplanes that have not been modified per Boeing Service Bulletin 757-54-0035. Therefore, no change is necessary in this regard to the AD.

Requests To Revise Inspection Method for Loose or Missing Fasteners

Two commenters request that a method of inspecting for loose or missing fasteners without the engine in place be specified. The commenters state that the inspection method specified in the supplemental NPRM is burdensome to accomplish with the engine in place.

We do not agree with the commenters' request. Since the manufacturer has not provided us with service information describing such a method of inspection, we have not reviewed and approved such an inspection method. However, under the provisions of paragraph (h) of the AD, we may approve requests for an alternative inspection method if data are submitted to substantiate that such an alternative inspection method would provide an acceptable level of safety.

Request for an Alternative Inspection Method

One commenter, the manufacturer, requests that a simple gap check be performed with a feeler or wire gage in lieu of the inspection in paragraph (c) of the supplemental NPRM. The commenter explains that this can be done with the strut still installed, which is described in Boeing Service Bulletin 757-54-0035, Revision 2. The commenter further recommends that a minimum gap of 0.030 inch be maintained between the middle gusset on the inboard side load fitting and the strut clevis lug.

We do not agree with permitting such an alternative method of inspection at this time, since the gap check has not been sufficiently defined for us to review and approve. However, under the provisions of paragraph (h) of the

AD, we may approve requests for an alternative inspection method if data are submitted to substantiate that such an alternative method would provide an acceptable level of safety.

Request To Use Alternative Method of Oversizing Holes

One commenter requests approval for using procedures to oversize holes specified in the Structural Repair Manual (SRM) in lieu of using Boeing-supplied tools specified in paragraph (g) of the supplemental NPRM. The commenter notes that there is a limited supply of those tools.

We do not agree with the commenter's request. In certain cases, operator supplied tools have contributed to unsafe conditions. However, under the provisions of paragraph (h) of the AD, we may approve requests for an alternative method of oversizing holes if data are submitted to substantiate that such a method to oversize holes would provide an acceptable level of safety.

Conclusion

After careful review of the available data, including the comments noted above, the FAA has determined that air safety and the public interest require the adoption of the rule with the changes previously described. The FAA has determined that these changes will neither increase the economic burden on any operator nor increase the scope of the AD.

Change to the Code of Federal Regulations

On July 10, 2002, the FAA issued a new version of 14 CFR part 39 (67 FR 47997, July 22, 2002), which governs the FAA's airworthiness directives system. The regulation now includes material that relates to altered products, special flight permits, and alternative methods of compliance. Because we have now included this material in part 39, only the office authorized to approve AMOCs is identified in each individual AD. However, for clarity and consistency in this AD, we have retained the language of the supplemental NPRM regarding that material.

Change to Labor Rate Estimate

We have reviewed the figures we have used over the past several years to calculate AD costs to operators. To account for various inflationary costs in the airline industry, we find it necessary to increase the labor rate used in these calculations from \$60 per work hour to \$65 per work hour. The cost impact information, below, reflects this increase in the specified hourly labor rate.

Cost Impact

There are approximately 394 airplanes of the affected design in the worldwide fleet. The FAA estimates that 176 airplanes of U.S. registry will be affected by this AD.

The modification that is currently required by AD 99-24-07 takes approximately 1,049 work hours per airplane to accomplish, at an average labor rate of \$65 per work hour. This work hour figure includes the time it will take to remove and reinstall the struts from the airplane as well as the time required to gain and close access to the adjacent wing structure. Based on these figures, the cost impact of the currently required modification on U.S. operators is estimated to be \$12,000,560, or \$68,185, per airplane.

This cost impact figure does not reflect the cost of the terminating actions described in the service bulletins listed in paragraph I.C., Table I, "Strut Improvement Bulletins," on page 7 of Revision 2 of Boeing Service Bulletin 757-54-0035, that are required to be accomplished prior to, or concurrently with, the modification of the nacelle strut and wing structure. Since some operators may have accomplished certain modifications on some or all of the airplanes in the fleet, while other operators may not have accomplished any of the modifications on any of the airplanes in the fleet, the FAA is unable to provide a reasonable estimate of the cost of accomplishing the terminating actions described in the service bulletins listed in Table I of the service bulletin.

It will take approximately 1 work hour per airplane to accomplish the new detailed inspection of the middle gusset, at an average labor rate of \$65 per work hour. Based on these figures, the cost impact of the inspection required by this AD is estimated to be \$11,440, or \$65 per airplane.

It will take approximately 8 work hours per airplane to accomplish the new fastener removal and eddy current inspection, at an average labor rate of \$65 per work hour. Based on these figures, the cost impact of the removal and inspection required by this AD is estimated to be \$91,520, or \$520 per airplane.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions

actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. Section 39.13 is amended by removing amendment 39-11431 (64 FR 66370, November 26, 1999), and by adding a new airworthiness directive (AD), amendment 39-13666, to read as follows:

2004-12-07 Boeing: Amendment 39-13666. Docket 2000-NM-376-AD. Supersedes AD 99-24-07, Amendment 39-11431.

Applicability: Model 757 series airplanes equipped with Rolls Royce RB211 engines, line numbers 1 through 735 inclusive; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance (AMOC) in accordance with paragraph (h)(1) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent fatigue cracking in primary strut structure and consequent reduced structural integrity of the strut, accomplish the following:

Restatement of Requirements of AD 99-24-07

Modification

(a) Modify the nacelle strut and wing structure according to Boeing Service Bulletin 757-54-0035, dated July 17, 1997; Revision 1, dated April 15, 1999; or Revision 2, dated June 13, 2002, at the later of the times specified in paragraph (a)(1) or (a)(2) of this AD. All of the terminating actions described in the service bulletins and listed in paragraph I.C., Table I, "Strut Improvement Bulletins," on page 6 of Boeing Service Bulletin 757-54-0035, on page 7 of Revision 1 of the service bulletin, and on Page 7 of Revision 2 of the service bulletin, as applicable, must be accomplished according to those service bulletins prior to, or concurrently with, the accomplishment of the modification of the nacelle strut and wing structure required by this paragraph. After the effective date of this AD, use only Revision 2 of the service bulletin.

(1) Prior to the accumulation of 37,500 total flight cycles, or prior to 20 years since the date of manufacture of the airplane, whichever occurs first.

(2) Within 3,000 flight cycles after January 3, 2000 (the effective date of AD 99-24-07, amendment 39-11431).

New Requirements of This AD

Inspections/Corrective Actions

(b) For airplanes on which the modification required by paragraph (a) of this AD has not been done according to Boeing Service Bulletin 757-54-0035, dated July 17, 1997: Before the accumulation of 15,000 total flight cycles, or within 6 months after the effective date of this AD, whichever is later, do a detailed inspection of the 20 aft bulkhead fasteners of the lower spar fitting for loose or missing fasteners, according to a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA. Before further flight, replace any loose or missing fasteners with new fasteners according to Boeing Service Bulletin 757-54-0035, Revision 1, dated April 15, 1999; or Revision 2, dated June 13, 2002, excluding Evaluation Form. Repeat the inspection

thereafter at intervals not to exceed 6,000 flight cycles or 36 months, whichever occurs first. Accomplishment of the actions required by paragraph (a) of this AD constitutes terminating action for the requirements of this paragraph.

Note 2: The 20 aft bulkhead fasteners are located in Panel 7 at Locations 36, 37, and 41. The number of fasteners at Location 37 has increased from 2 to 8 fasteners. Figure 30 of Boeing Service Bulletin 757-54-0035, Revision 2, dated June 13, 2002, illustrates the location of the fasteners.

Note 3: For the purposes of this AD, a detailed inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

(c) For airplanes on which the modification required by paragraph (a) of this AD has been done according to Boeing Service Bulletin 757-54-0035, dated July 17, 1997: Within 15,000 flight cycles after doing the modification required by paragraph (a) of this AD, or within 3 years after the effective date of this AD, whichever is later; do a one-time detailed inspection of the middle gusset of the inboard side load fitting for proper alignment, according to Part II of the Accomplishment Instructions of Boeing Service Bulletin 757-54-0035, Revision 1, dated April 15, 1999; or Revision 2, dated June 13, 2002, excluding Evaluation Form. If the gusset is not aligned properly, before further flight, machine the gusset to the specified angle according to the service bulletin.

(d) Before further flight after doing paragraph (c) of this AD, do the actions required by paragraphs (d)(1) and (d)(2) of this AD.

(1) Remove the aft bulkhead fasteners of the lower spar fitting and do a one-time eddy current inspection of those fastener holes for cracking, according to Part V of the Accomplishment Instructions of Boeing Service Bulletin 757-54-0035, Revision 1, dated April 15, 1999; or Revision 2, dated June 13, 2002, excluding Evaluation Form.

(2) Do a detailed inspection of the 8 fasteners of the lower spar fitting for loose or missing fasteners, according to a method approved by the Manager, Seattle ACO. Before further flight, replace any loose or missing fasteners with new fasteners according to Boeing Service Bulletin 757-54-0035, Revision 1, dated April 15, 1999; or Revision 2, dated June 13, 2002, excluding Evaluation Form.

Note 4: The 8 fasteners are located in Panel 7 at Location 37. The number of fasteners at Location 37 has increased from 2 to 8 fasteners. Figure 30 of Boeing Service Bulletin 757-54-0035, Revision 2, dated June 13, 2002, excluding Evaluation Form, illustrates the location of the fasteners.

(e) If any cracking is found during any inspection required by this AD: Before

further flight, repair according to a method approved by the Manager, Seattle ACO; or according to data meeting the type certification basis of the airplane approved by a Boeing Company Designated Engineering Representative who has been authorized by the Manager, Seattle ACO, to make such findings. For a repair method to be approved by the Manager, Seattle ACO, as required by this paragraph, the Manager's approval letter must specifically reference this AD.

(f) If no cracking is found during the inspection required by paragraph (d) of this AD, before further flight, increase the diameter of the fastener holes and install new fasteners according to Boeing Service Bulletin 757-54-0035, Revision 2, dated June 13, 2002, excluding Evaluation Form.

(g) Except as identified in Figures 3 and 5 of the Accomplishment Instructions of Boeing Service Bulletin 757-54-0035, Revision 2, dated June 13, 2002, excluding Evaluation Form, the actions must be done using Boeing-supplied tools.

Alternative Methods of Compliance

(h)(1) An AMOC or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle ACO. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

(2) AMOCs, approved previously in accordance with AD 99-24-07, amendment 39-11431, are approved as AMOCs with paragraph (a) of this AD.

Note 5: Information concerning the existence of approved AMOCs with this AD, if any, may be obtained from the Seattle ACO.

Special Flight Permit

(i) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(j) Unless otherwise specified, the actions shall be done in accordance with Boeing Service Bulletin 757-54-0035, Revision 1, dated April 15, 1999; or Boeing Service Bulletin 757-54-0035, Revision 2, dated June 13, 2002. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Boeing Commercial Airplanes, P.O. Box 3707, Seattle, Washington 98124-2207. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Effective Date

(k) This amendment becomes effective on July 21, 2004.

Issued in Renton, Washington, on May 25, 2004.

Kalene C. Yanamura,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 04-13144 Filed 6-15-04; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 71

[Docket No. FAA-2004-17496; Airspace Docket No. 04-AAL-04]

Establishment of Class E Airspace; Allakaket, AK

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: This action establishes Class E airspace at Allakaket, AK to provide adequate controlled airspace to contain aircraft executing two new Standard Instrument Approach Procedures (SIAP) and a new Textual Departure Procedure. This Rule results in new Class E airspace upward from 700 feet (ft.) and 1,200 feet above the surface at Allakaket, AK.

DATES: *Effective Date:* 0901 UTC, September 30, 2004.

FOR FURTHER INFORMATION CONTACT: Jesse Patterson, AAL-538G, Federal Aviation Administration, 222 West 7th Avenue, Box 14, Anchorage, AK 99513-7587; telephone number (907) 271-5898; fax: (907) 271-2850; email: Jesse.ctr.Patterson@faa.gov. Internet address: <http://www.alaska.faa.gov/at>.

SUPPLEMENTARY INFORMATION:

History

On Monday, April 19, 2004, the FAA proposed to revise part 71 of the Federal Aviation Regulations (14 CFR part 71) to create new Class E airspace upward from 700 ft. and 1,200 ft. above the surface at Allakaket, AK (69 FR 20835). The action was proposed in order to add Class E airspace sufficient in size to contain aircraft while executing two new Standard Instrument Approach Procedures and a new Textual Departure Procedure for the Allakaket Airport. The new approaches are Area Navigation-Global Positioning System (RNAV GPS) Runway (RWY) 5, original and (2) RNAV (GPS) Runway 23, original. New Class E controlled airspace extending upward from 700

feet and 1,200 feet above the surface in the Allakaket Airport area is established by this action. Interested parties were invited to participate in this rulemaking proceeding by submitting written comments on the proposal to the FAA. No public comments have been received, thus, the rule is adopted as proposed.

The area will be depicted on aeronautical charts for pilot reference. The coordinates for this airspace docket are based on North American Datum 83. The Class E airspace areas designated as 700/1200 foot transition areas are published in paragraph 6005 of FAA Order 7400.9L, *Airspace Designations and Reporting Points*, dated September 2, 2003, and effective September 16, 2003, which is incorporated by reference in 14 CFR 71.1. The Class E airspace designation listed in this document will be revoked and revised subsequently in the Order.

The Rule

This revision to 14 CFR part 71 establishes Class E airspace at Allakaket, Alaska. This additional Class E airspace was created to accommodate aircraft executing two new SIAPs and will be depicted on aeronautical charts for pilot reference. The intended effect of this rule is to provide adequate controlled airspace for IFR operations at Allakaket Airport, Allakaket, Alaska.

The FAA has determined that this regulation only involves an established body of technical regulations for which frequent and routine amendments are necessary to keep them operationally current. It, therefore—(1) is not a “significant regulatory action” under Executive Order 12866; (2) is not a “significant rule” under DOT Regulatory Policies and Procedures (44 FR 11034; February 26, 1979); and (3) does not warrant preparation of a regulatory evaluation as the anticipated impact is so minimal. Since this a routine matter that will only affect air traffic procedures and air navigation, it is certified that this rule will not have a significant economic impact on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 71

Airspace, Incorporation by reference, Navigation (air).

Adoption of the Amendment

■ In consideration of the foregoing, the Federal Aviation Administration amends 14 CFR part 71 as follows: